

AD 683763

**THRUST COEFFICIENTS,
THRUST DEFLECTION ANGLES, AND
NONDIMENSIONAL MOMENTS
FOR NOZZLES WITH OBLIQUE EXIT PLANES**

22-125619-1

Best Available Copy

MAR 17 1969
LIBRARY
N.A.

THE BOEING COMPANY ■ SEATTLE, WASHINGTON

Unclassified

Security Classification

DOCUMENT CONTROL DATA - R&D

(Security classification of title, body of abstract and indexing annotation must be entered when the overall report is classified).

1. ORIGINATING ACTIVITY (Corporate author) The Boeing Company Seattle, Washington		2a. REPORT SECURITY CLASSIFICATION Unclassified	
		2b. GROUP	
3. REPORT TITLE THRUST COEFFICIENTS, THRUST DEFLECTION ANGLES, AND NONDIMENSIONAL MOMENTS FOR NOZZLES WITH OBLIQUE EXIT PLANES			
4. DESCRIPTIVE NOTES (Types of report and inclusive dates)			
5. AUTHORS (First name, middle initial, last name) William W. Phillips, Ross A. Fiedler, Robert G. Hoptcroft			
6. REPORT DATE March, 1968		7a. TOTAL NO. OF PAGES 543	7b. NO. OF REFS
8a. CONTRACT OR GRANT NO. b. c. d.		9a. ORIGINATOR'S REPORT NUMBERS D2-125619-1	
		9b. OTHER REPORT NO(S) (Any other numbers that may be assigned this report)	
10. DISTRIBUTION STATEMENT Distribution of this document is unlimited.			
11. SUPPLEMENTARY NOTES		12. SPONSORING MILITARY ACTIVITY	
13. ABSTRACT → Nozzles with oblique exit planes, scarfed nozzles, are encountered in aerospace applications such as reaction control and thrust reversal systems. This document provides thrust coefficient, thrust deflection angle, and nondimensional turning moment data for scarfed conical nozzles on machine-generated graphs. Based on one-dimensional isentropic flow of a perfect gas, the data cover specific heat ratios from 1.18 to 1.40, nozzle half-angles from 5 to 25 degrees, scarf angles from 10 to 60 degrees, area ratios at the upstream scarf point from 1.5 to 15, and ratios of chamber pressure to ambient pressure from 20 to infinity (vacuum).			

Unclassified

Security Classification

DATE	BY	REVIEWED
6/10		
DIST. BY ROUTE AND/OR SPECIAL DIST.		

D2-125619-1

THRUST COEFFICIENTS, THRUST DEFLECTION ANGLES,
AND NONDIMENSIONAL MOMENTS FOR NOZZLES
WITH OBLIQUE EXIT PLANES

March 1968

MAR 17 1969

RECEIVED
DEFENSE RESEARCH AND ENGINEERING
AGENCY

Missile and Information Systems Division
Aerospace Group
THE BOEING COMPANY
Seattle, Washington

CONTENTS

	<u>Page</u>
PREFACE	v
SYMBOLS	vi
ANALYSIS	viii
Assumptions and Limitations	viii
Equations	viii
Computational Accuracy	ix
EXPLANATION OF GRAPHS	x
Use of Graphs	x
Plotting Accuracy	x
Sequence of Graphs	x
GRAPHS	1

LIST OF GRAPHS

Graphs of thrust coefficient, deflection angle, and nondimensional moment are listed below.

γ	α	<u>Figures</u>
1.18	5	1-27
	10	28-54
	15	55-81
	20	82-108
	25	109-135
1.20	5	136-162
	10	163-189
	15	190-216
	20	217-243
	25	244-270
1.22	5	271-297
	10	298-324
	15	325-351
	20	352-378
	25	379-405
1.40	5	406-432
	10	433-459
	15	460-486
	20	487-513
	25	514-540

PREFACE

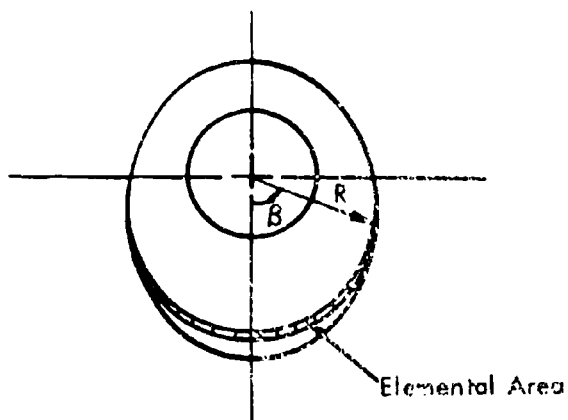
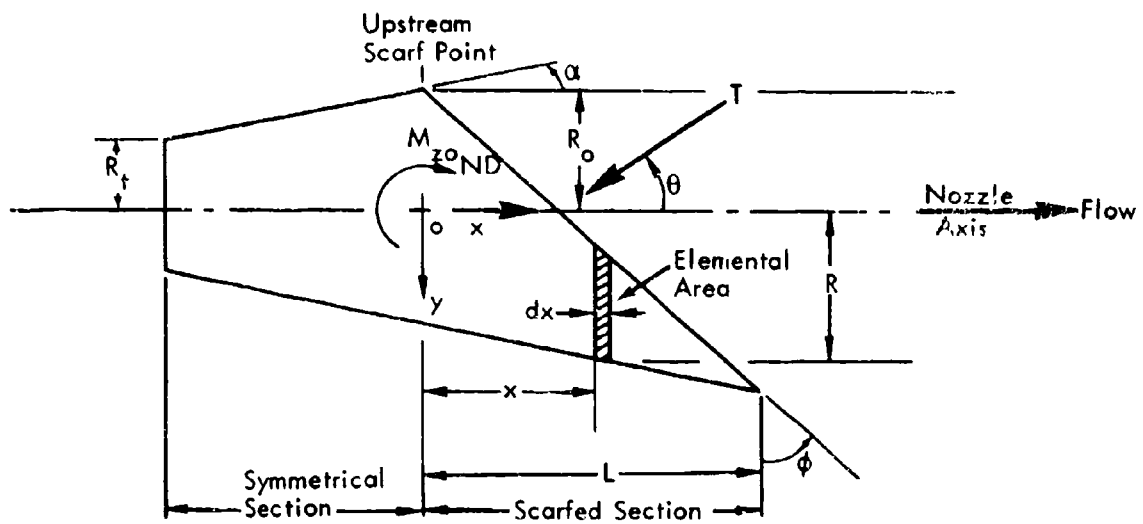
Nozzles with oblique exit planes, scarfed nozzles, are encountered in aerospace applications such as reaction control and thrust reversal systems. This document provides thrust coefficient, thrust deflection angle, and nondimensional turning moment data for scarfed conical nozzles on machine-generated graphs. Based on one-dimensional isentropic flow of a perfect gas, the data cover specific heat ratios from 1.18 to 1.40, nozzle half-angles from 5 to 25 degrees, scarf angles from 10 to 60 degrees, area ratios at the upstream scarf point from 1.5 to 15, and ratios of chamber pressure to ambient pressure from 20 to infinity (vacuum).

The data presented herein were developed by the Rocket Propulsion Research Staff of the Missile and Information Systems Division of The Boeing Company. An original analysis by Robert Hopcroft was extended in scope and programmed for computation by Ross Fiedler and William Phillips. Mike Whitaker programmed the machine plotting. William Phillips wrote the text and coordinated printing.

SYMBOLS

(See Figures A and B on Page vii)

A_t	Nozzle throat area
$C_{F_{sc}}$	Total thrust coefficient
C_{F_o}	Thrust coefficient for symmetrical section of nozzle (corrected for flow divergence)
C_{T_x}	Total thrust coefficient in x-direction
C_{T_y}	Total thrust coefficient in y-direction
F_x	Thrust produced by scarfed section of nozzle in x-direction
F_y	Thrust produced by scarfed section of nozzle in y-direction
L	Length of scarfed section of nozzle
$M_{z_{oND}}$	Nondimensional moment produced about z-axis at point o (positive in clockwise direction)
P	Pressure at any point inside scarfed section
P_a	Ambient pressure
P_c	Chamber pressure
R	Radius of nozzle at any point inside scarfed section
R_o	Radius of nozzle at upstream scarf point
R_t	Nozzle throat radius
T	Total thrust produced by the nozzle
T_x	Total thrust in x-direction
T_y	Total thrust in y-direction
x	Axial distance inside nozzle from point o
y	Distance inside nozzle perpendicular to axis
α	Nozzle half-angle
β	Direction of radius vector R
γ	Specific heat ratio
θ	Thrust deflection angle
ϕ	Nozzle scarf angle



ANALYSIS

ASSUMPTIONS AND LIMITATIONS

- 1) The flow is accelerated from stagnation conditions.
- 2) The gases obey the perfect gas laws.
- 3) The flow is one-dimensional, steady, and isentropic.
- 4) The momentum term of the thrust equation is corrected for divergence of the gas flow.
- 5) The data provided exclude those cases where:
 - a) Expansion fans originating from the upstream scarf point would intersect the opposite wall of the nozzle;
 - b) Flow separation would occur inside the nozzle.

EQUATIONS

The scarfed nozzle thrust coefficient, deflection angle, and nondimensional moment are defined as follows:

$$C_{F_{sc}} = \frac{T}{P_c A_t}$$

$$\theta = \tan^{-1} \left(\frac{T_y}{T_x} \right)$$

$$M_{zo_{ND}} = \frac{\int_0^L x \, dF_y + \int_0^L y \, dF_x}{P_c R_t^3}$$

The thrust coefficient and deflection angle are calculated from:

$$C_{F_{sc}} = \sqrt{C_{T_x}^2 + C_{T_y}^2}$$

$$\theta = \tan^{-1} \left(\frac{C_{T_y}}{C_{T_x}} \right)$$

where,

$$C_{T_x} = \frac{2 \tan \alpha}{\pi} \int_0^{L/R_t} \left(\frac{P}{P_c} - \frac{P_a}{P_c} \right) \frac{R}{R_t} \sin \beta d\left(\frac{x}{R_t}\right) + C_{F_o}$$

$$C_{T_y} = \frac{2}{\pi} \int_0^{L/R_t} \left(\frac{P}{P_c} - \frac{P_a}{P_c} \right) \frac{R}{R_t} \sin \beta d\left(\frac{x}{R_t}\right)$$

The nondimensional moment is calculated from:

$$M_{z_{o_{ND}}} = 2 \tan \alpha \int_0^{L/R_t} \left(\frac{P}{P_c} - \frac{P_a}{P_c} \right) \left(\frac{R}{R_t} \right)^2 \sin \beta d\left(\frac{x}{R_t}\right) \\ + 2 \int_0^{L/R_t} \left(\frac{P}{P_c} - \frac{P_a}{P_c} \right) \left(\frac{R}{R_t} \right) \sin \beta \frac{x}{R_t} d\left(\frac{x}{R_t}\right)$$

The moment axis is perpendicular to the nozzle plane of symmetry at a point on the nozzle axis opposite the upstream scarf point (point o in the diagram on Page vii). The pressure ratio, P/P_c , is determined from one-dimensional isentropic flow theory. The parameters R/R_t , L/R_t , and β are expressed in terms of the nozzle half-angle, scarf angle, and initial area ratio.

COMPUTATIONAL ACCURACY

The implicit relationship between pressure ratio and area ratio requires that an iterative computation be carried out to determine the pressure ratio at the upstream scarf point and the pressure profile in the scarfed section of the nozzle, which is required for numerical integration of the forces. The number of increments in the integration procedure has been selected to ensure an accuracy of 0.006% in the calculations.

EXPLANATION OF GRAPHS

USE OF GRAPHS

The plots of the performance parameters as functions of the ratio of chamber pressure to ambient pressure are discontinued at the maximum *calculated* value of initial expansion ratio (less than 15.0) for which separation does not occur downstream in the scarfed section. The performance parameters were calculated for each half-unit of expansion ratio from 1.5 to 15. Separation criteria from the accumulation of data presented* are employed to provide limits to the calculated data for initial expansion ratios of less than 15.

With initial expansion ratios equal to or greater than 1.5, the minimum for which data are calculated, expansion waves originating from the upstream scarf point will not intersect the opposite wall (thus influencing the pressure distribution on the nozzle wall), provided the scarf angle is not greater than 60 degrees for nozzle half-angles of 5 degrees or more. Any extrapolation of the curves below a scarf-point area ratio of 1.5 must be checked for this condition.

Linear interpolation may be used to obtain values of the performance parameters for low intermediate values of scarf angle and nozzle half-angle without incurring errors greater than 1%. For values of scarf angle greater than 40 degrees and nozzle half-angle greater than 20 degrees, errors of up to 10% may be incurred.

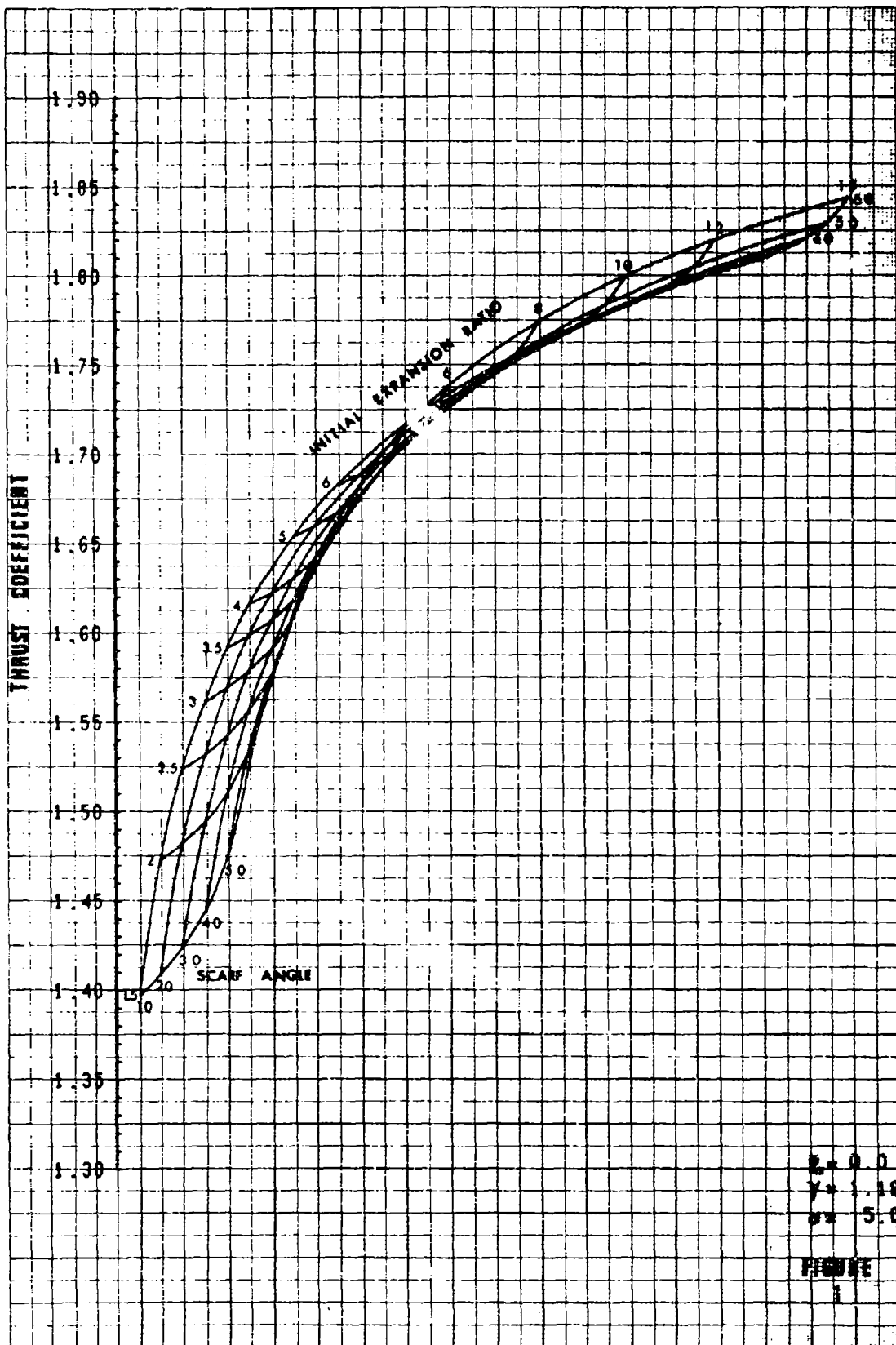
PLOTTING ACCURACY

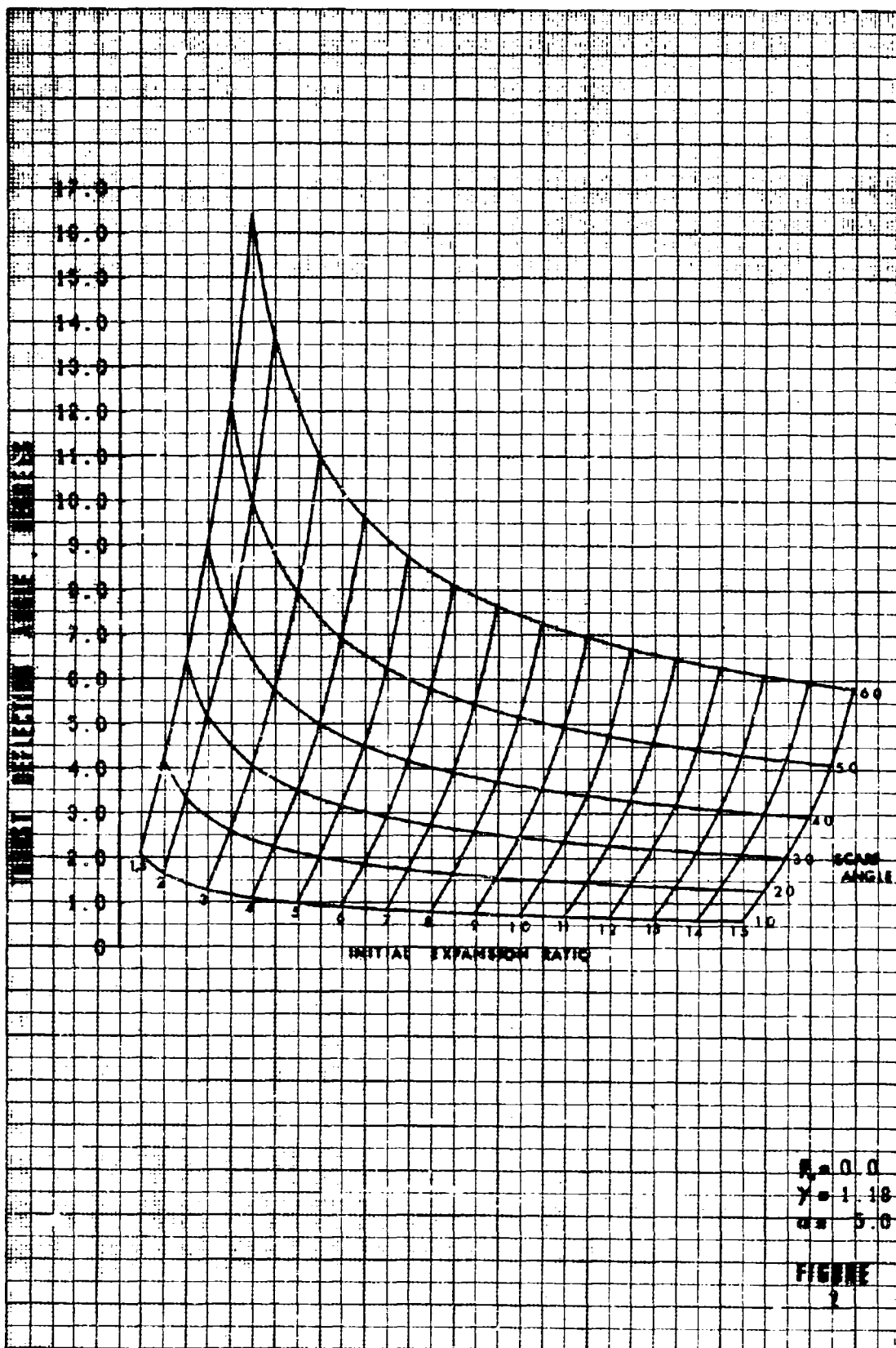
These graphs were machine-plotted with an accuracy of 0.5% compared to the computed data.

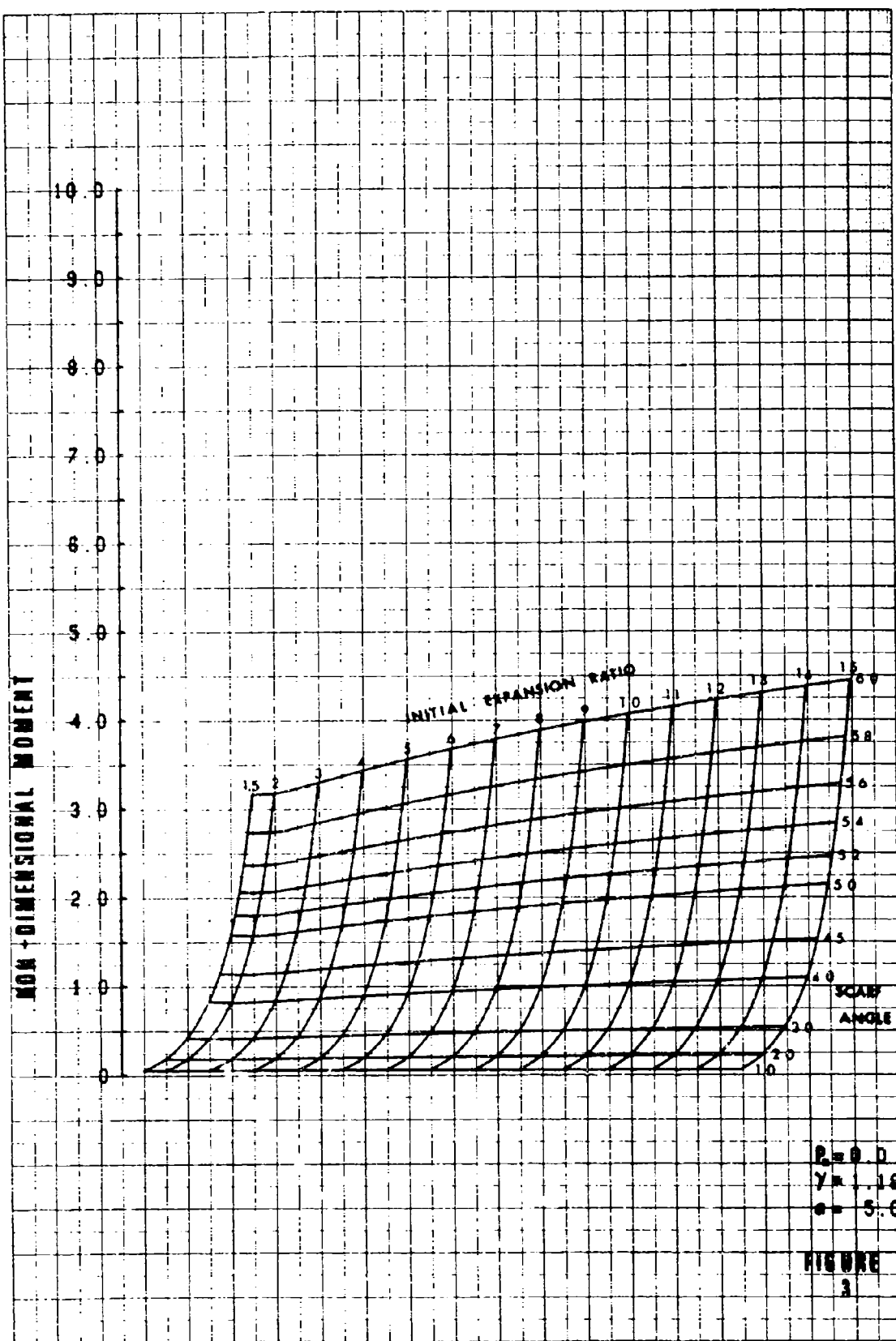
SEQUENCE OF GRAPHS

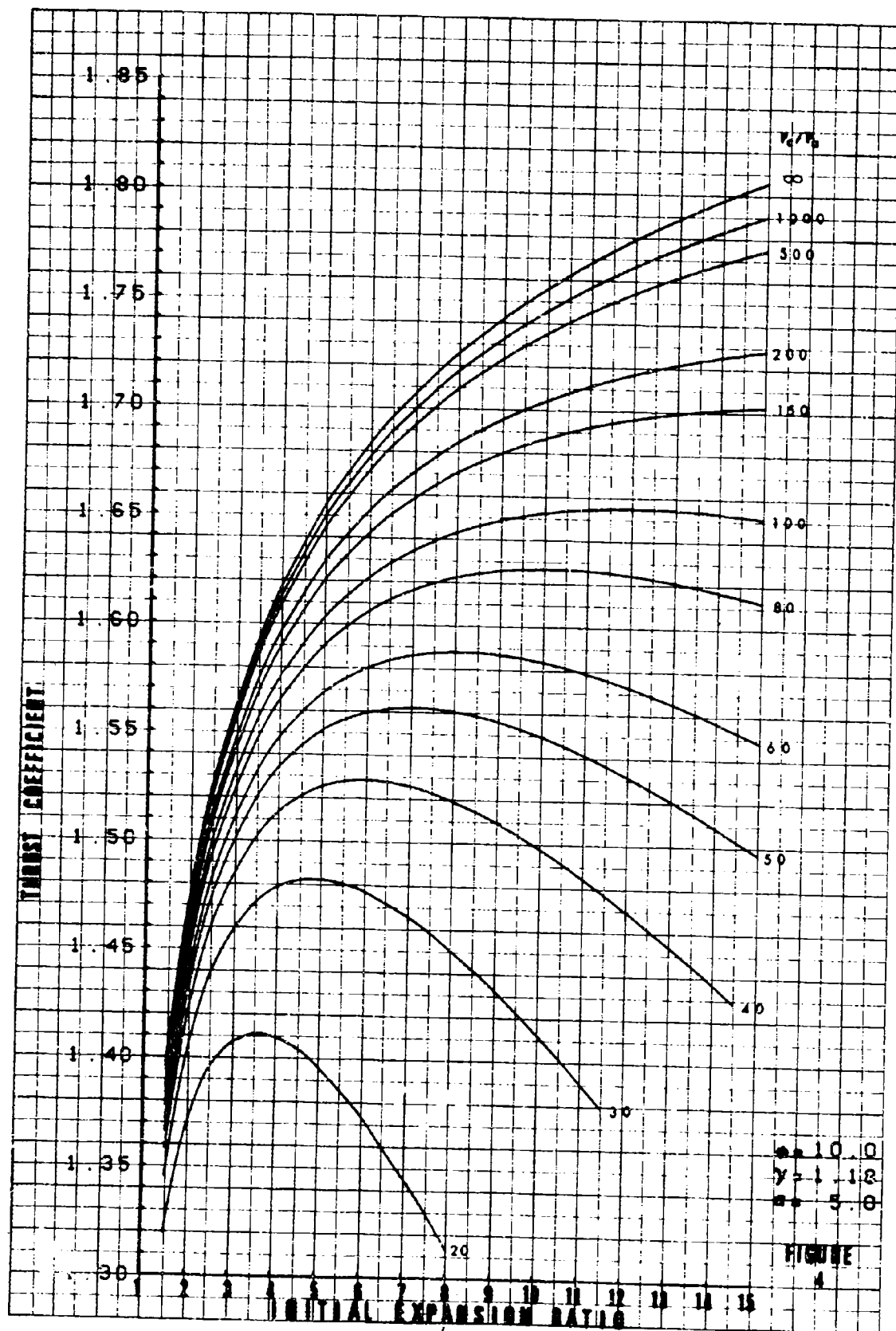
The graphs are divided into four sections, one for each value of specific heat ratio of 1.18, 1.20, 1.22, and 1.40. For each specific heat ratio there are five values of nozzle half-angle: 5, 10, 15, 20, and 25 degrees. For each nozzle half-angle there are 27 graphs, of which the first three are carpet plots of each of the performance parameters as functions of initial area ratio and scarf angle for vacuum conditions. On the following 24 pages, the graphs of thrust coefficient, deflection angle, and moment as functions of initial area ratio and ambient pressure are arranged together for each value of scarf angle: 10, 20, 30, 40, 45, 50, 55, and 60 degrees.

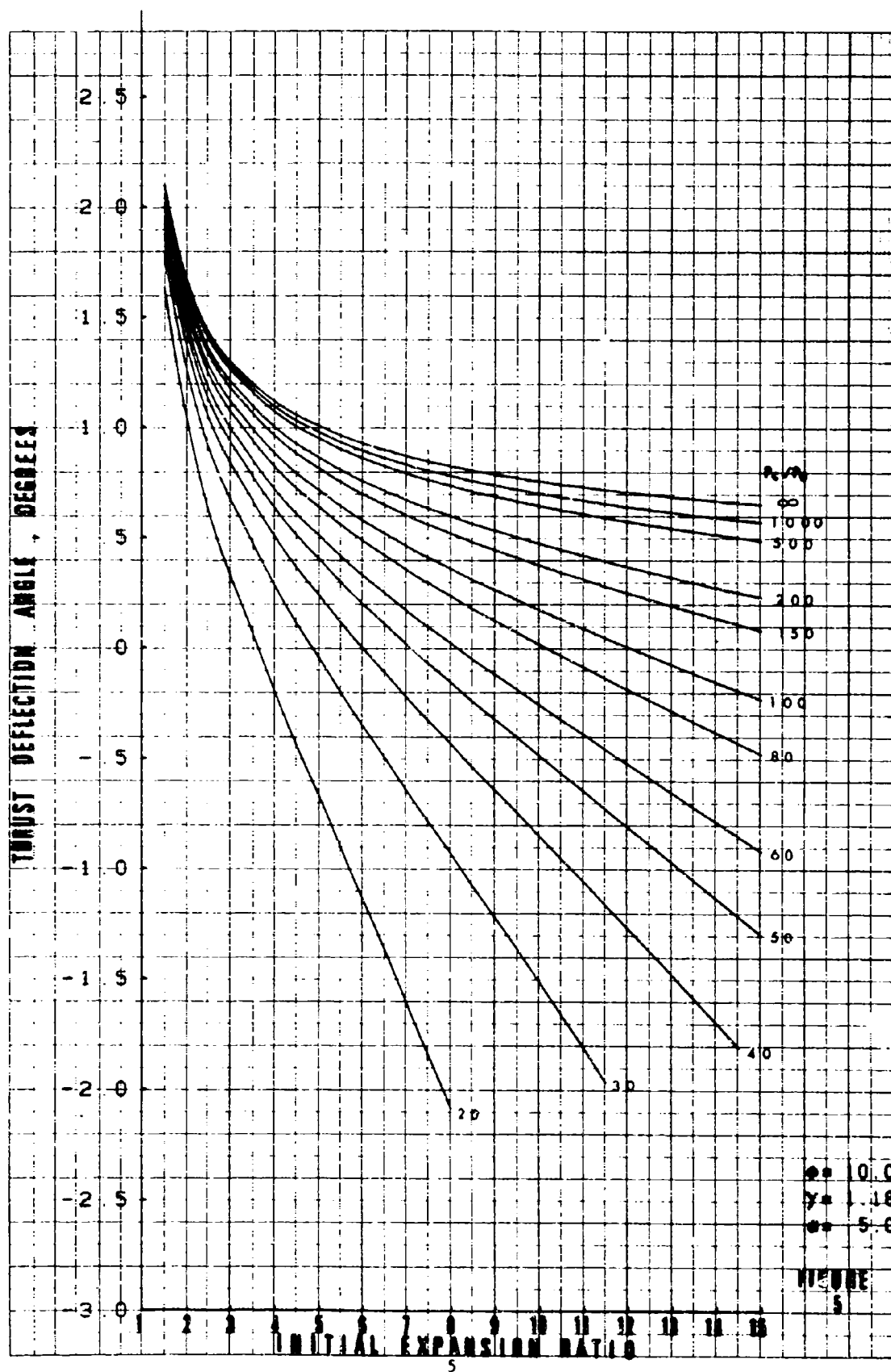
* Arens, M., and E. Spiegler, "Shock-Induced Boundary Layer Separation in Overexpanded Conical Exhaust Nozzles," *AIAA Journal*, March 1963.

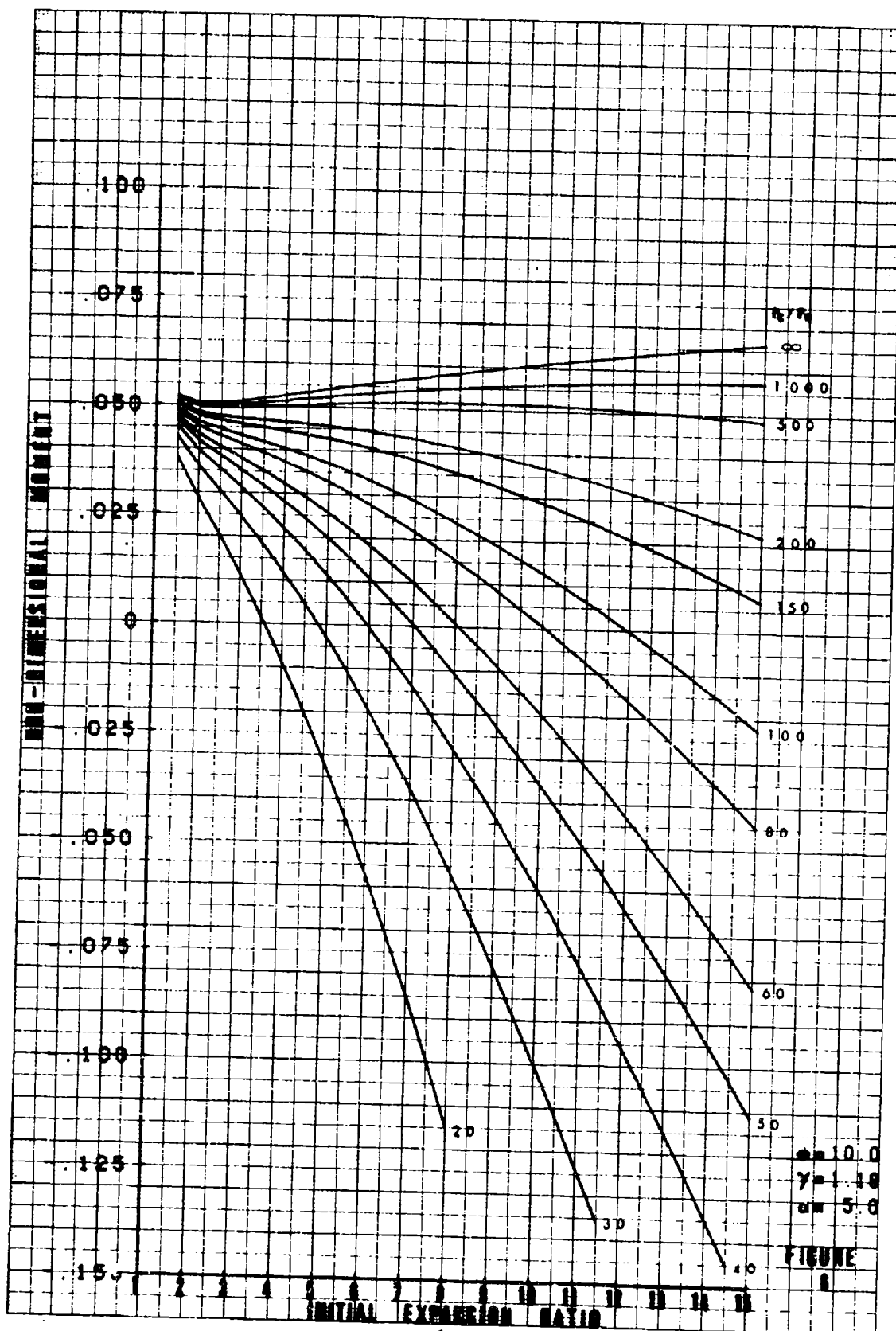












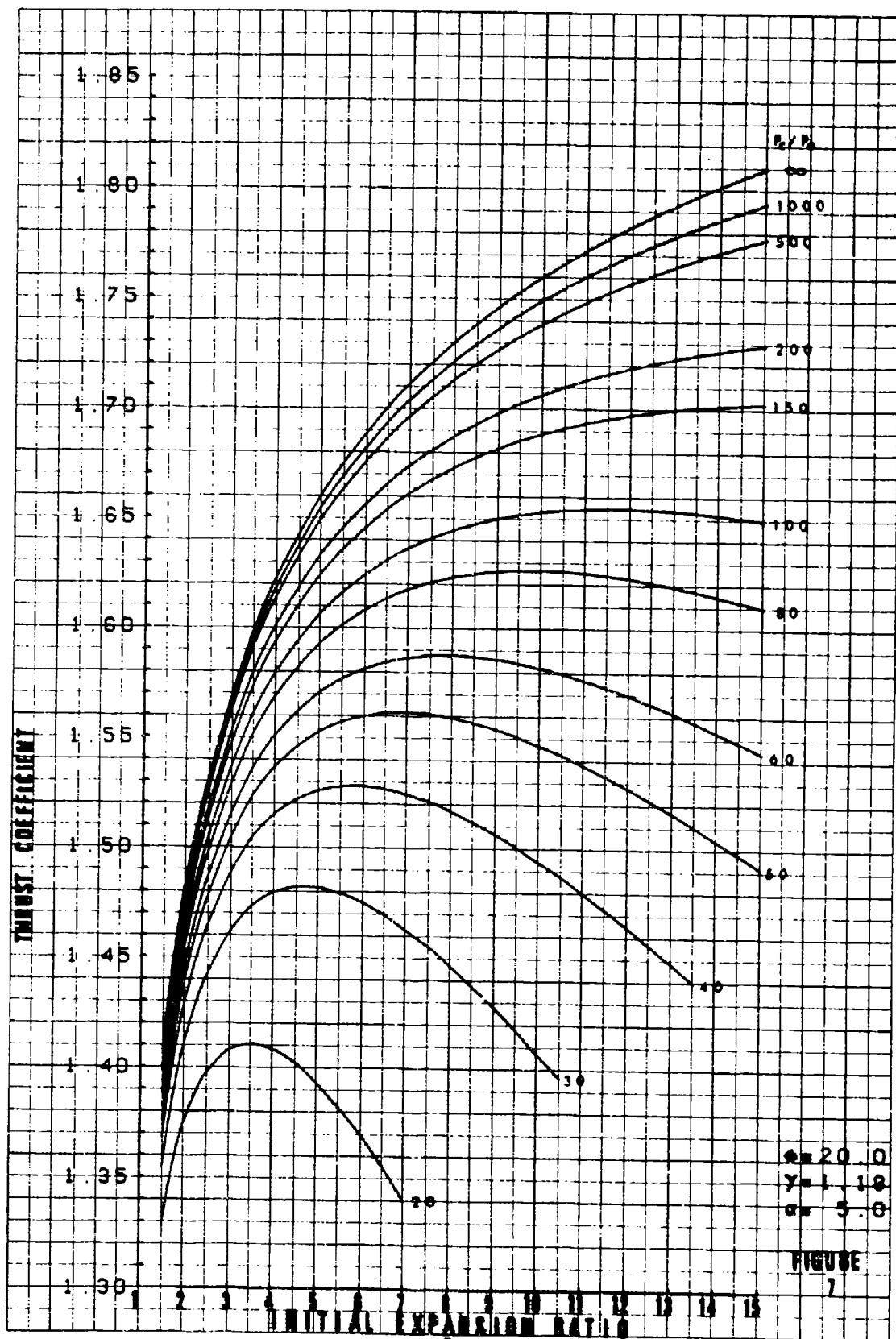
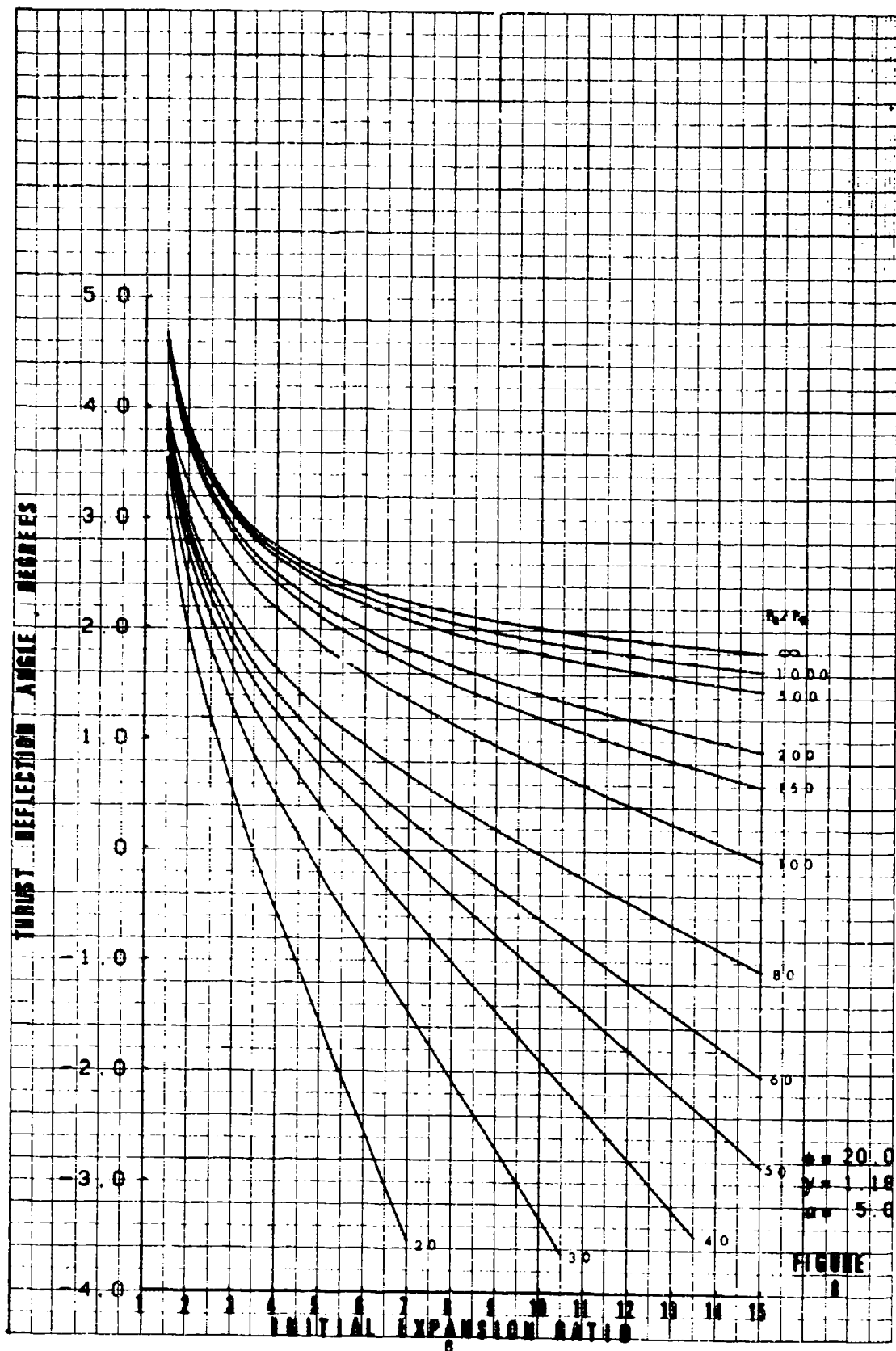
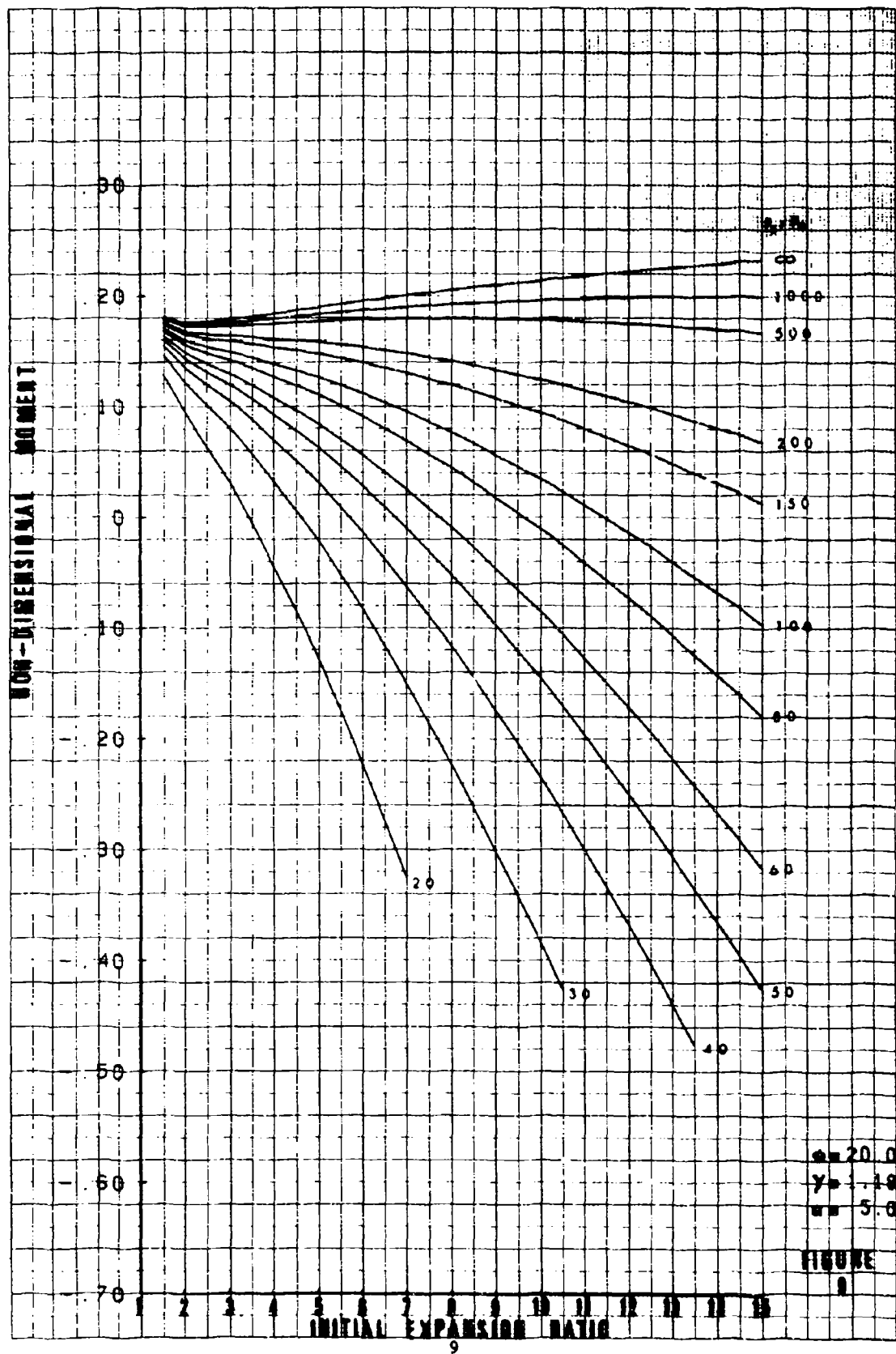
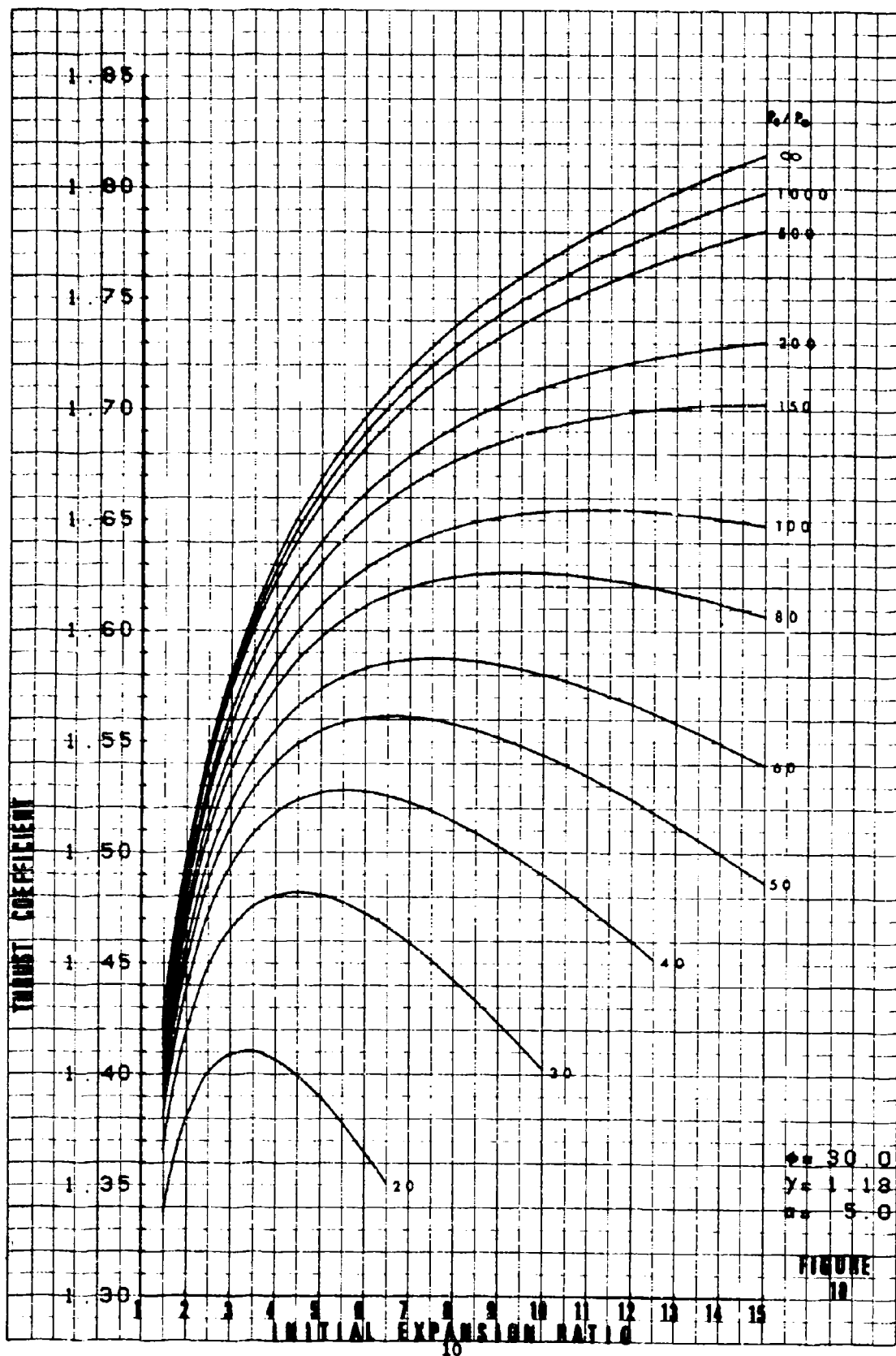


FIGURE 7







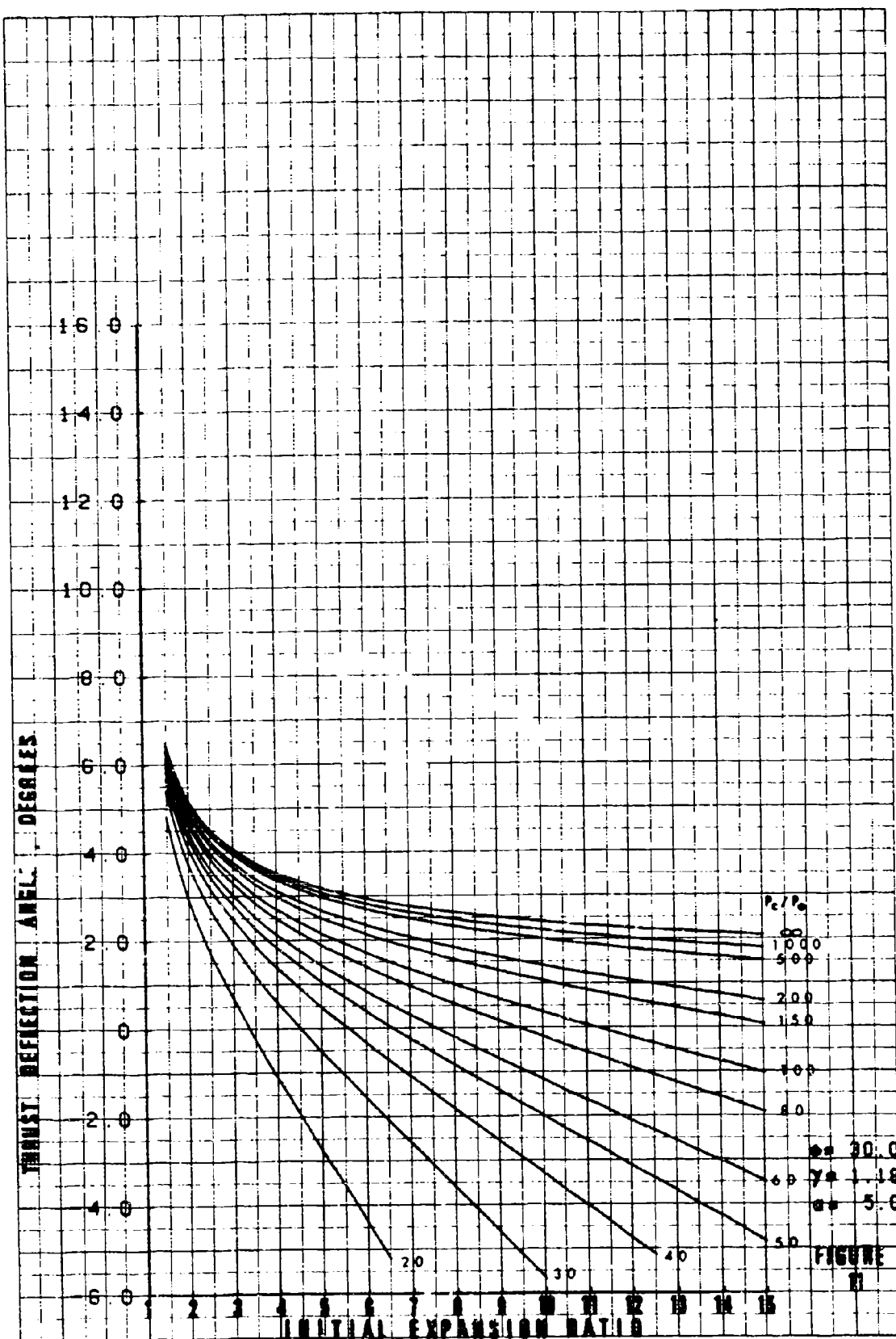


FIGURE 11

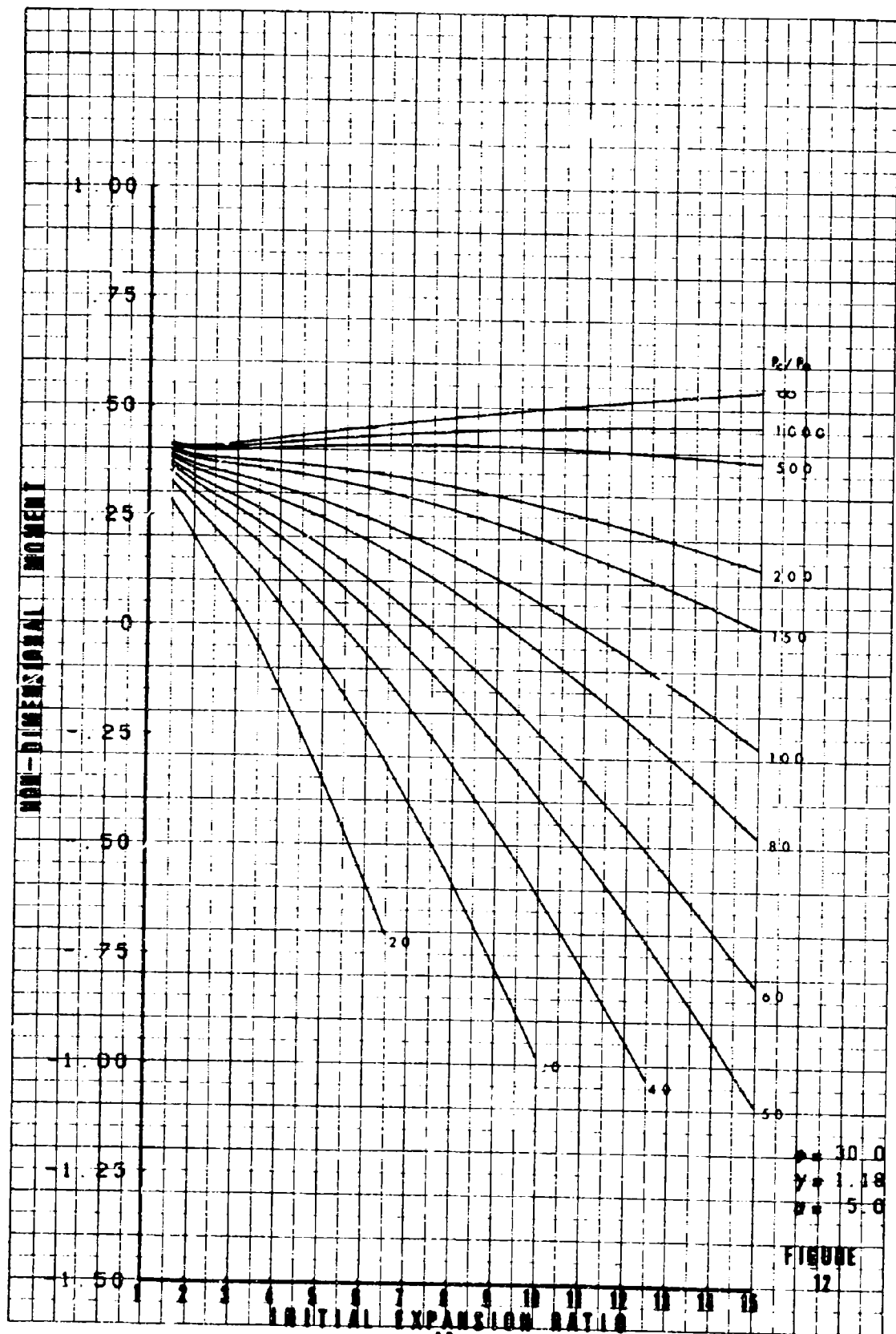
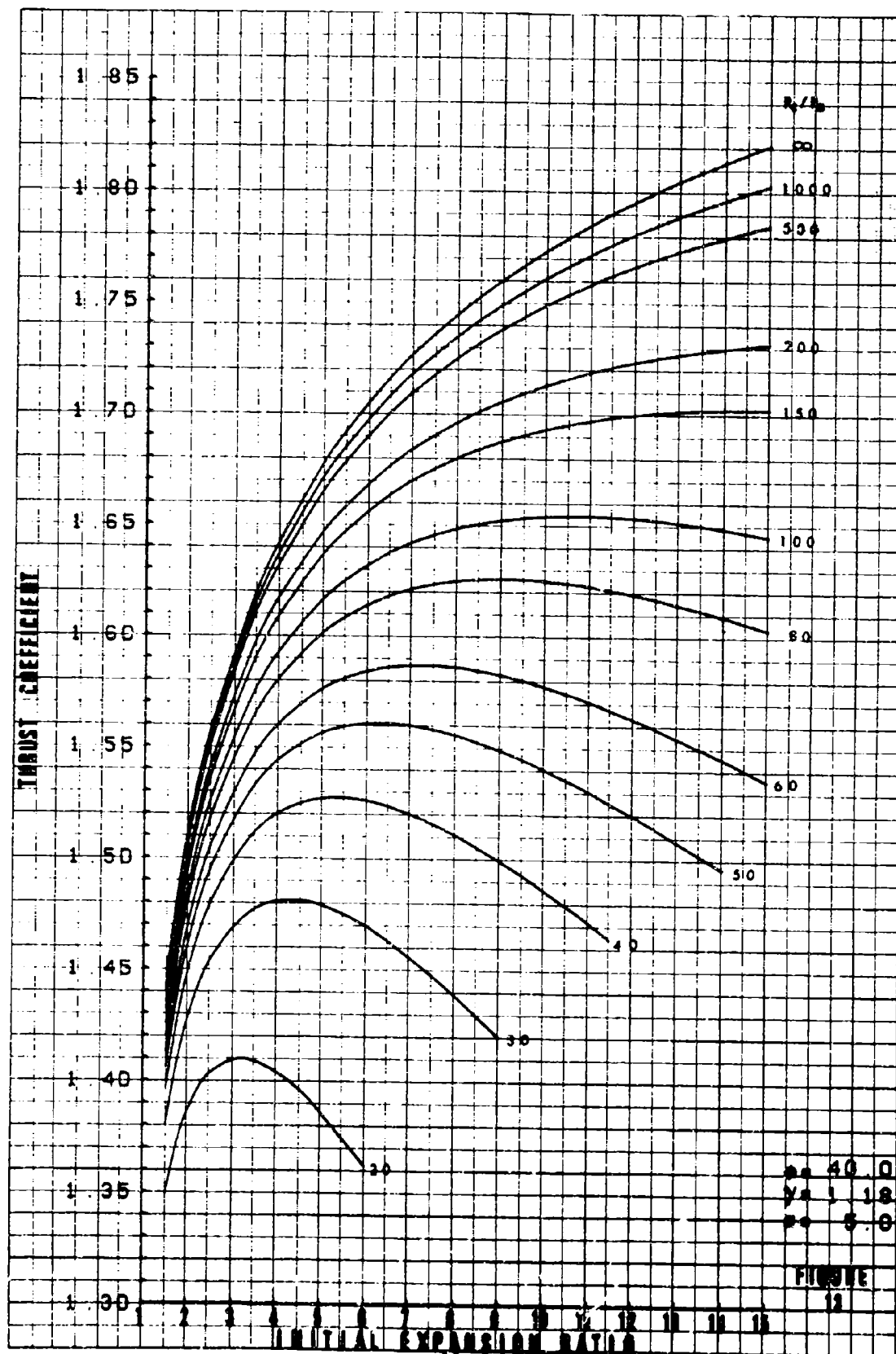


FIGURE 12



$\gamma = 40.0$
 $\gamma = 1.8$
 $\gamma = 5.0$

FIGURE

12

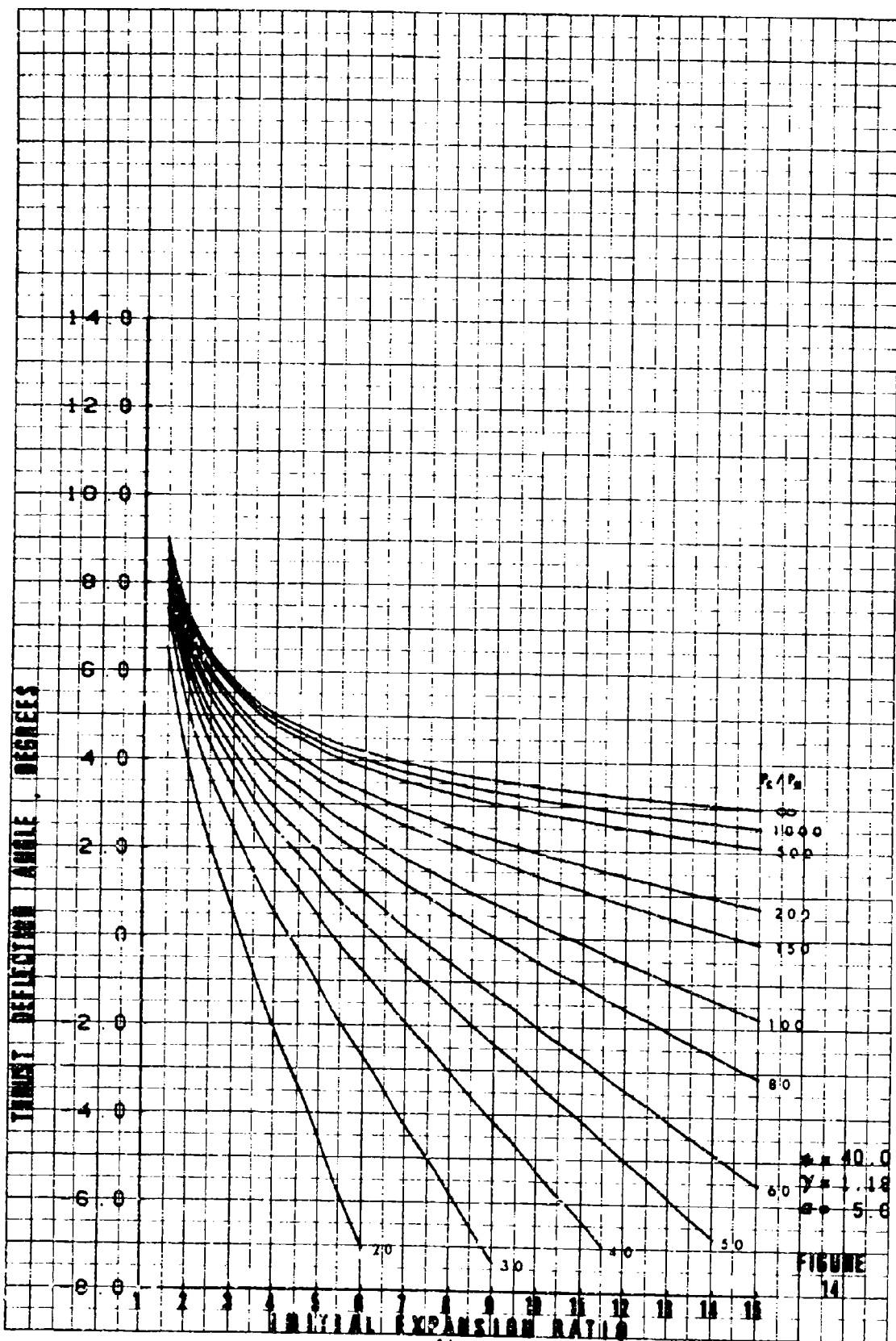
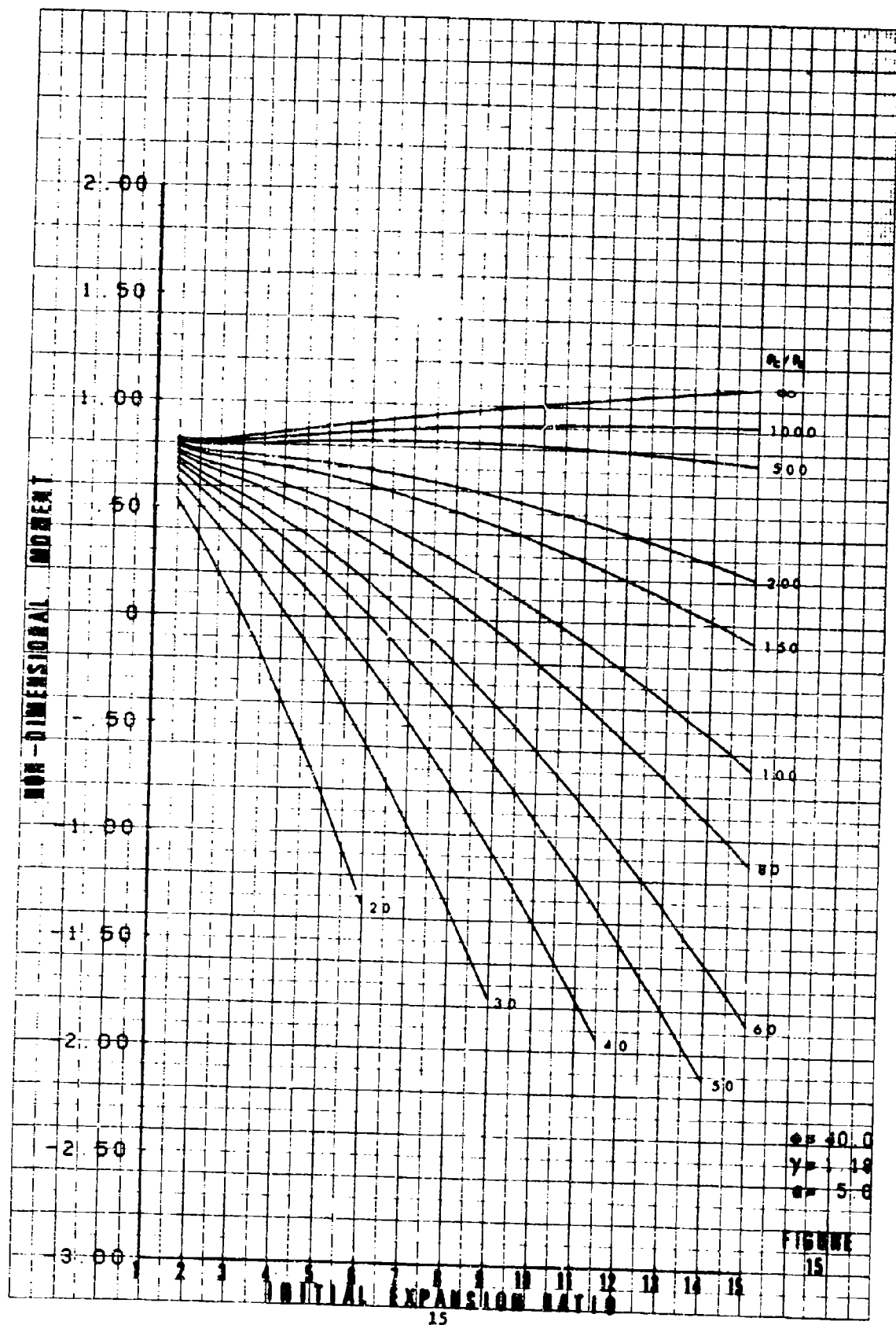
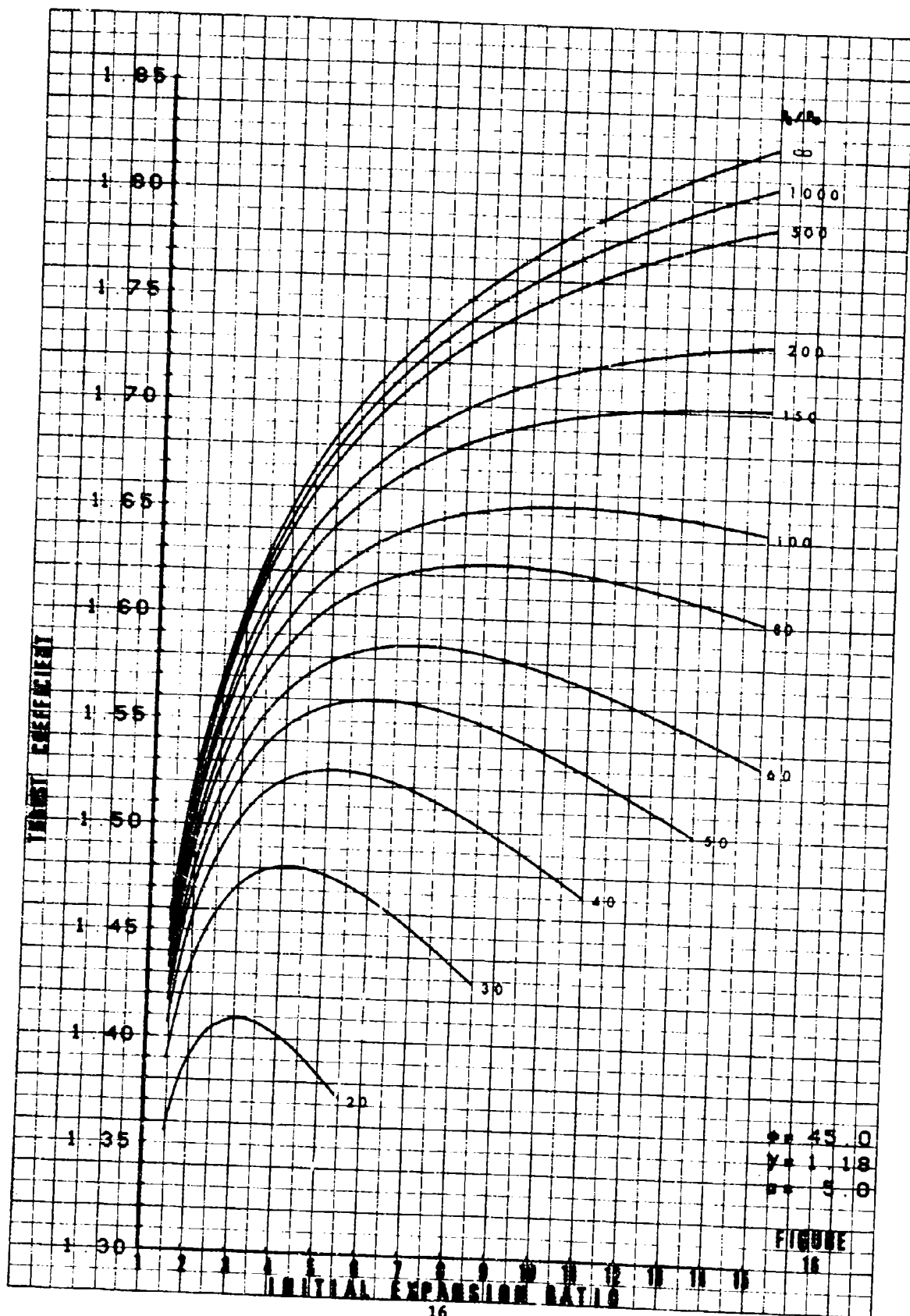


FIGURE 14





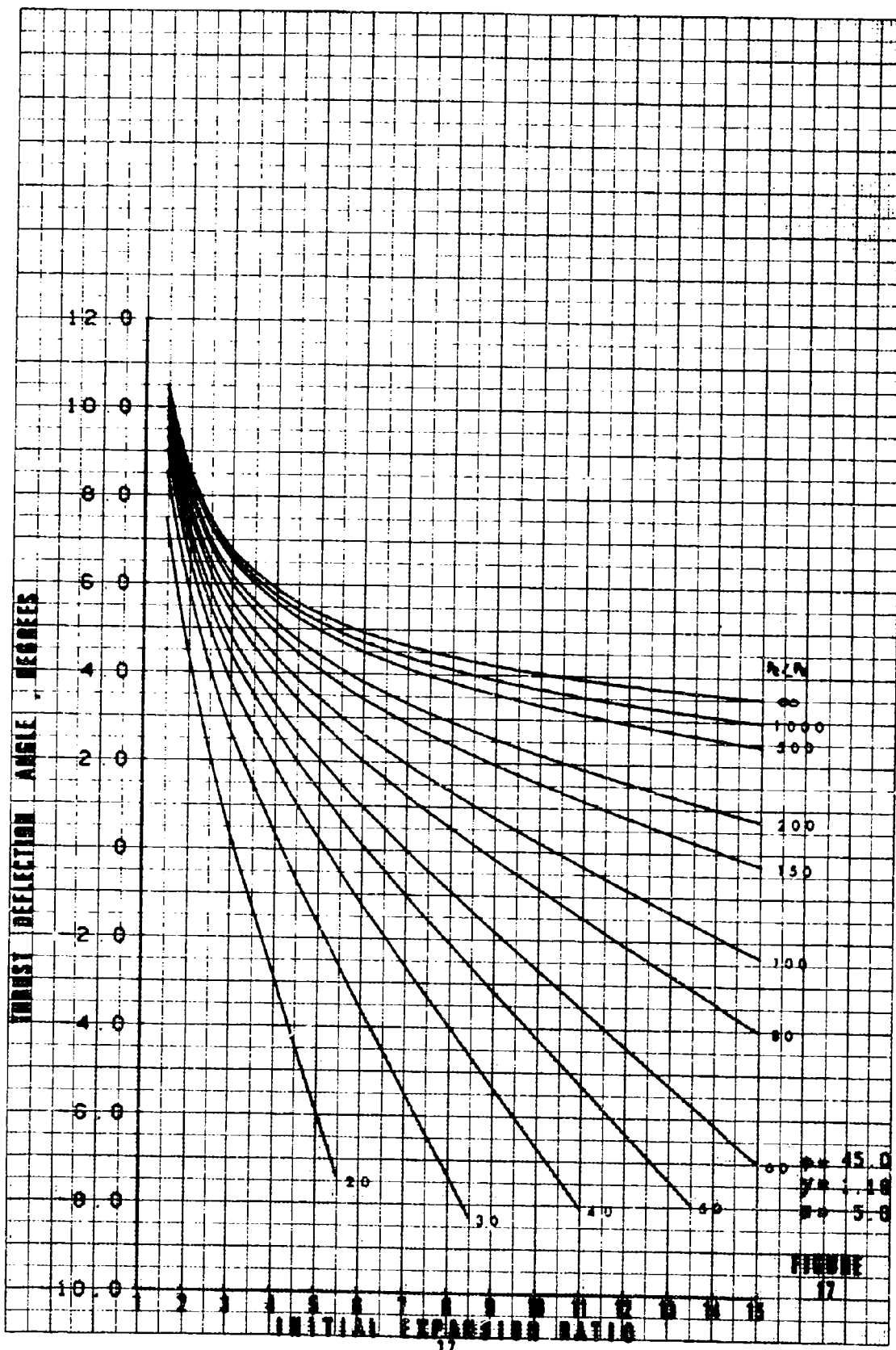
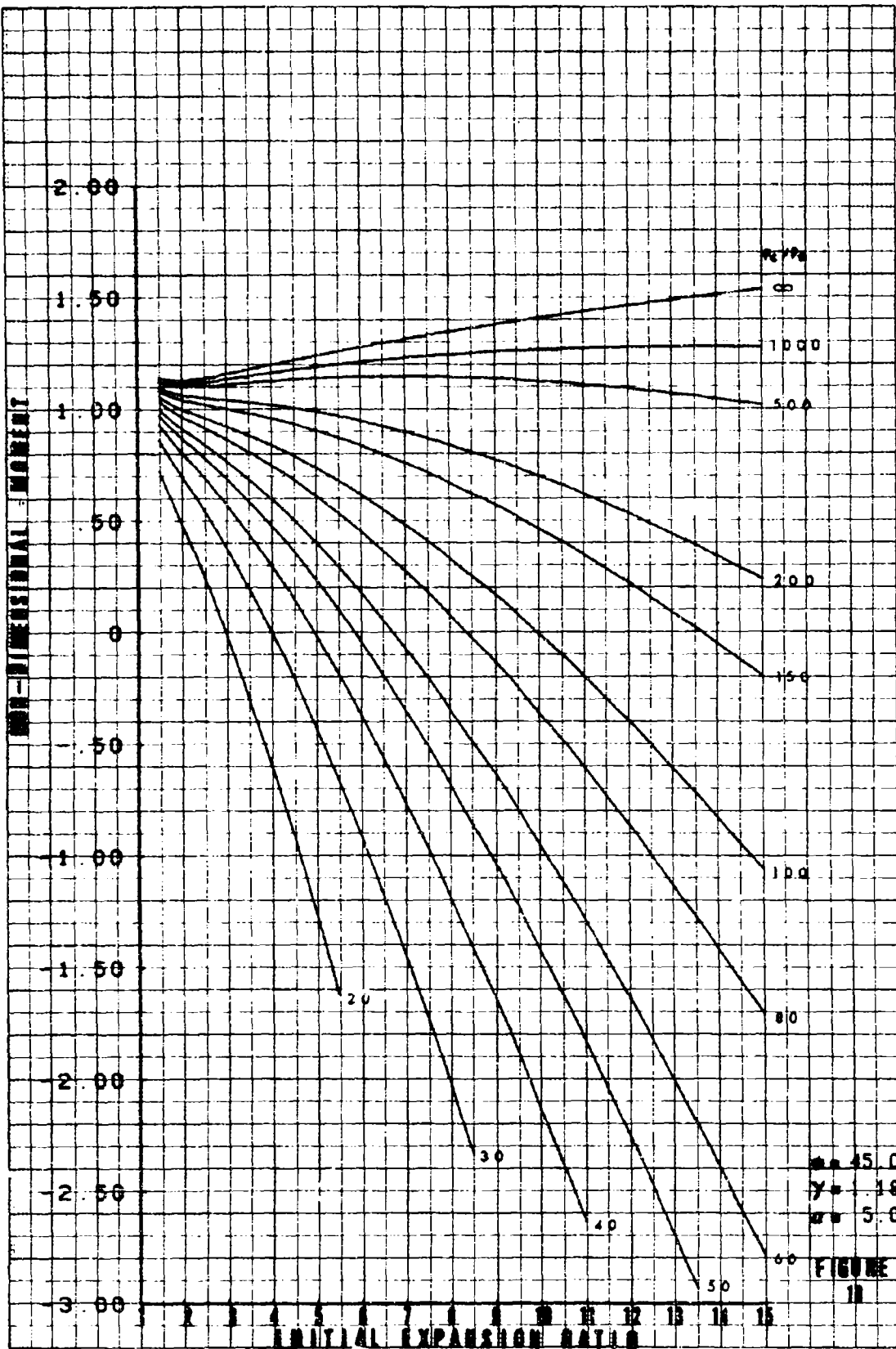
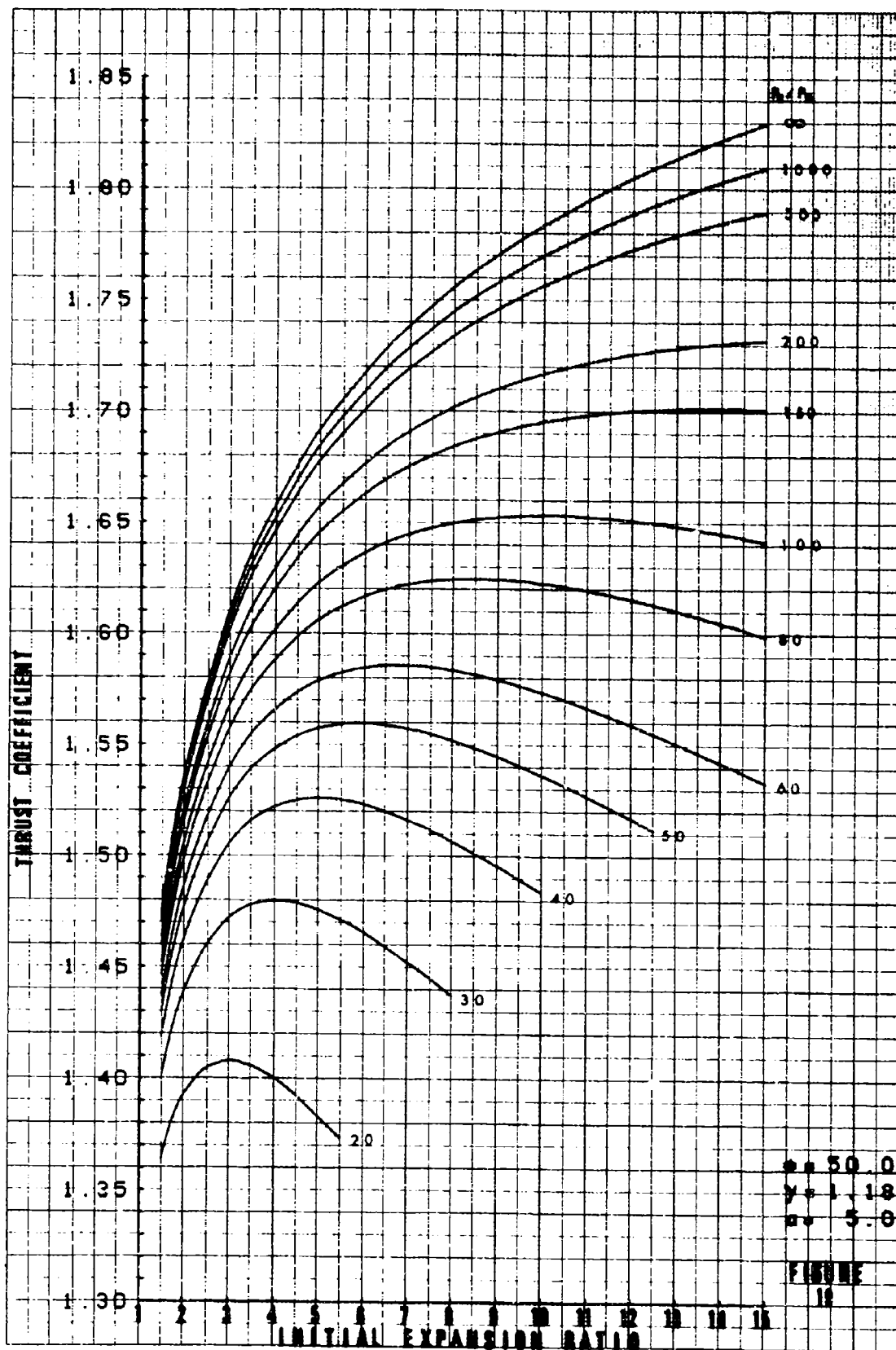
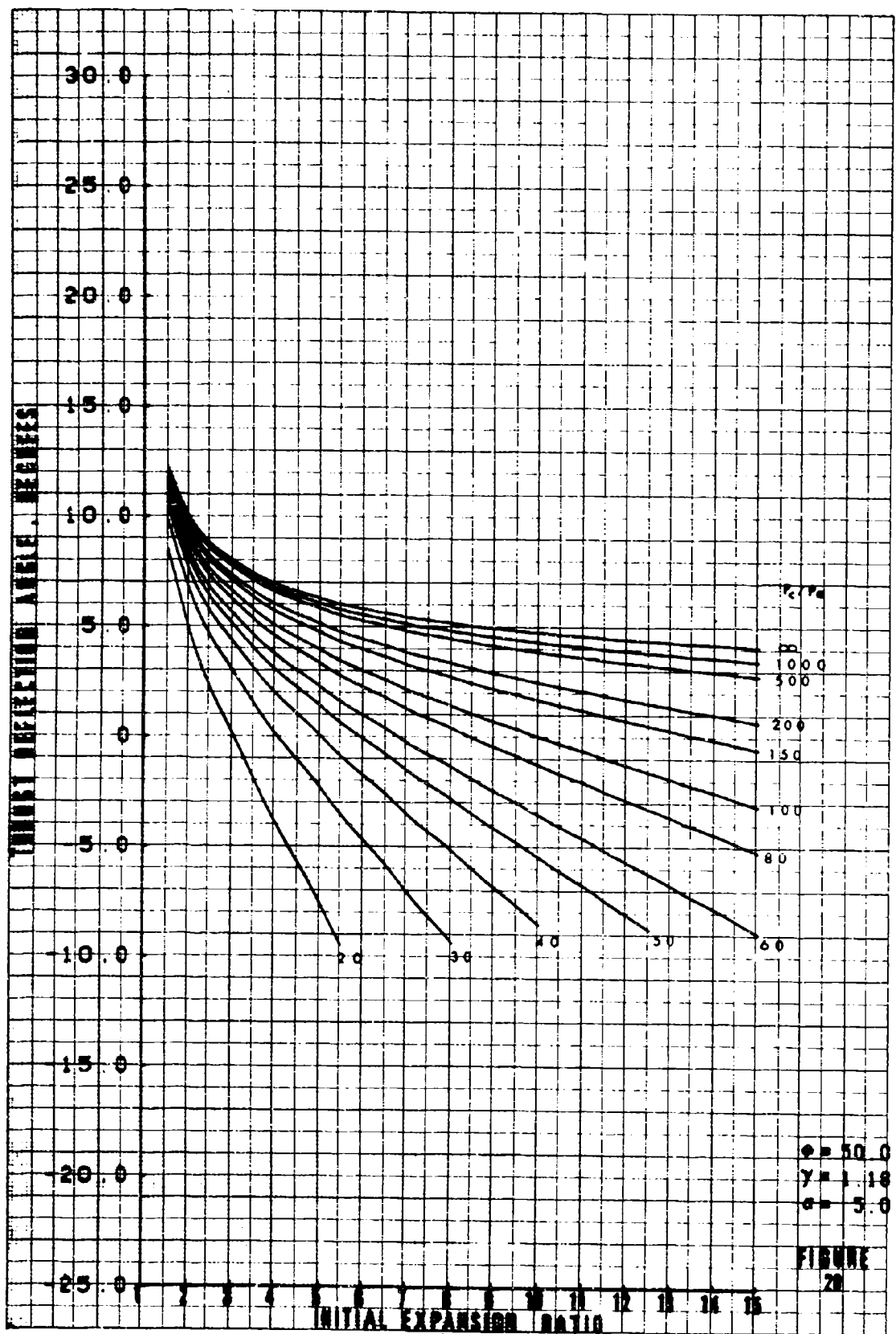
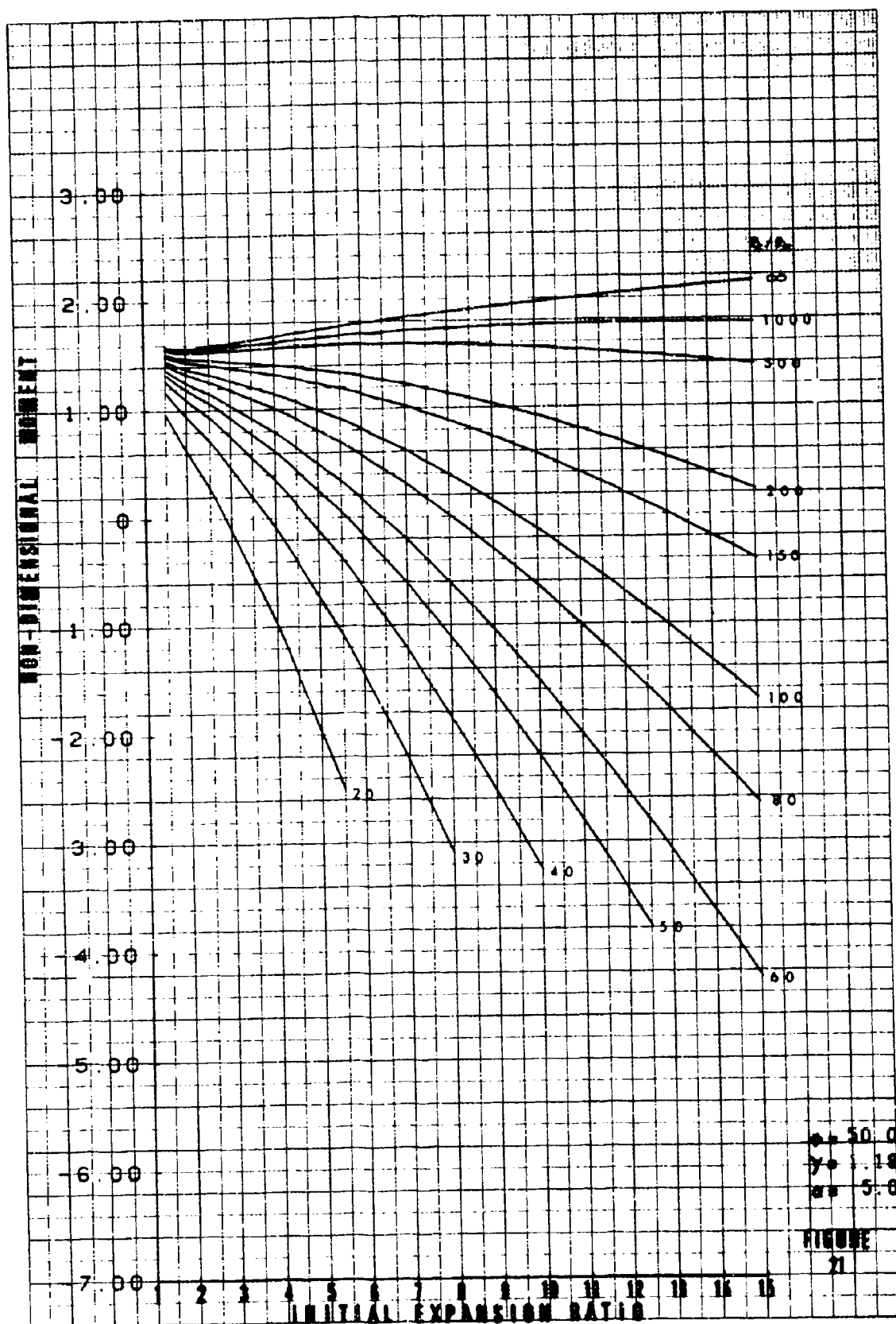


FIGURE 17



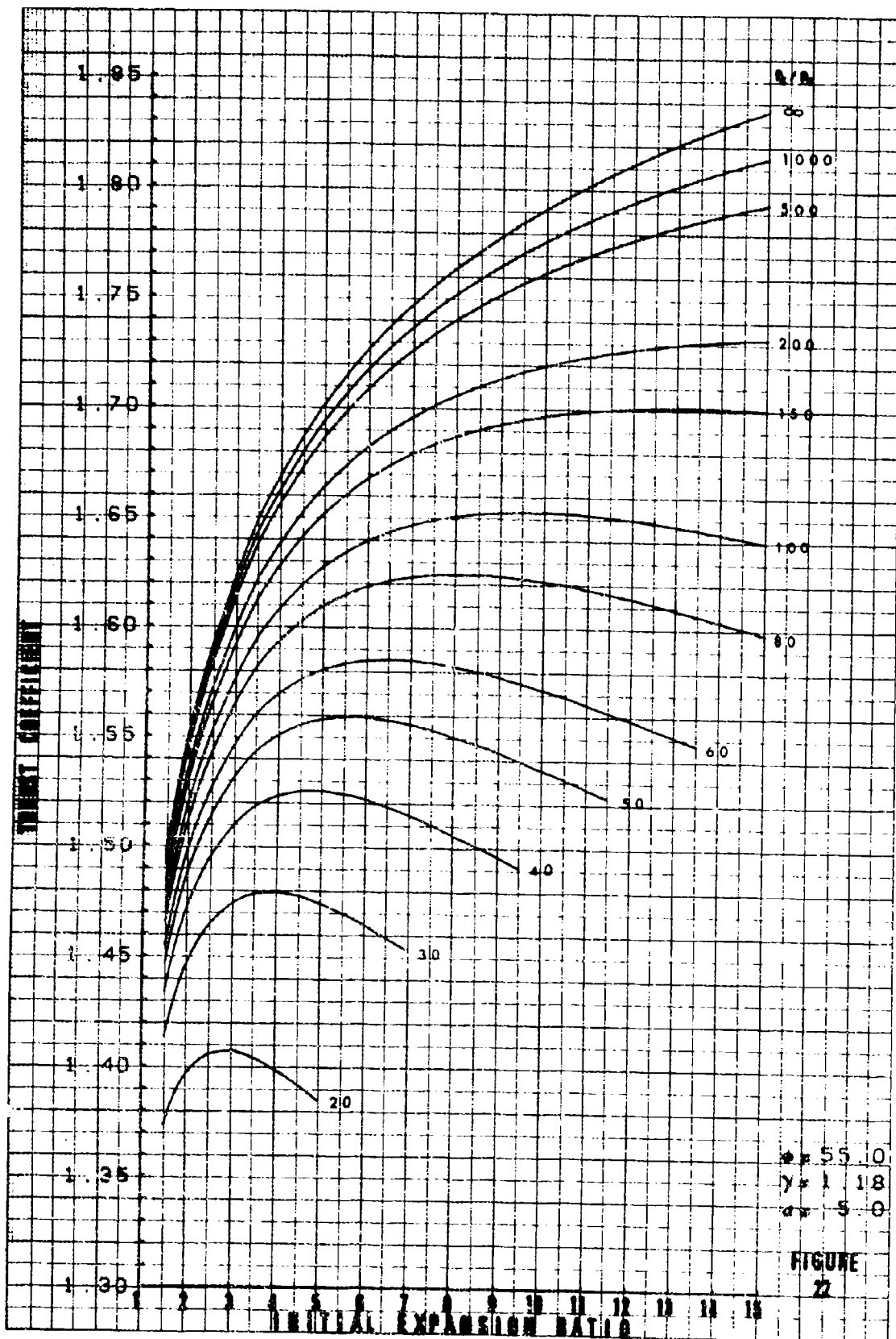






• 50.0
 • 1.0
 • 5.0

FIGURE 21



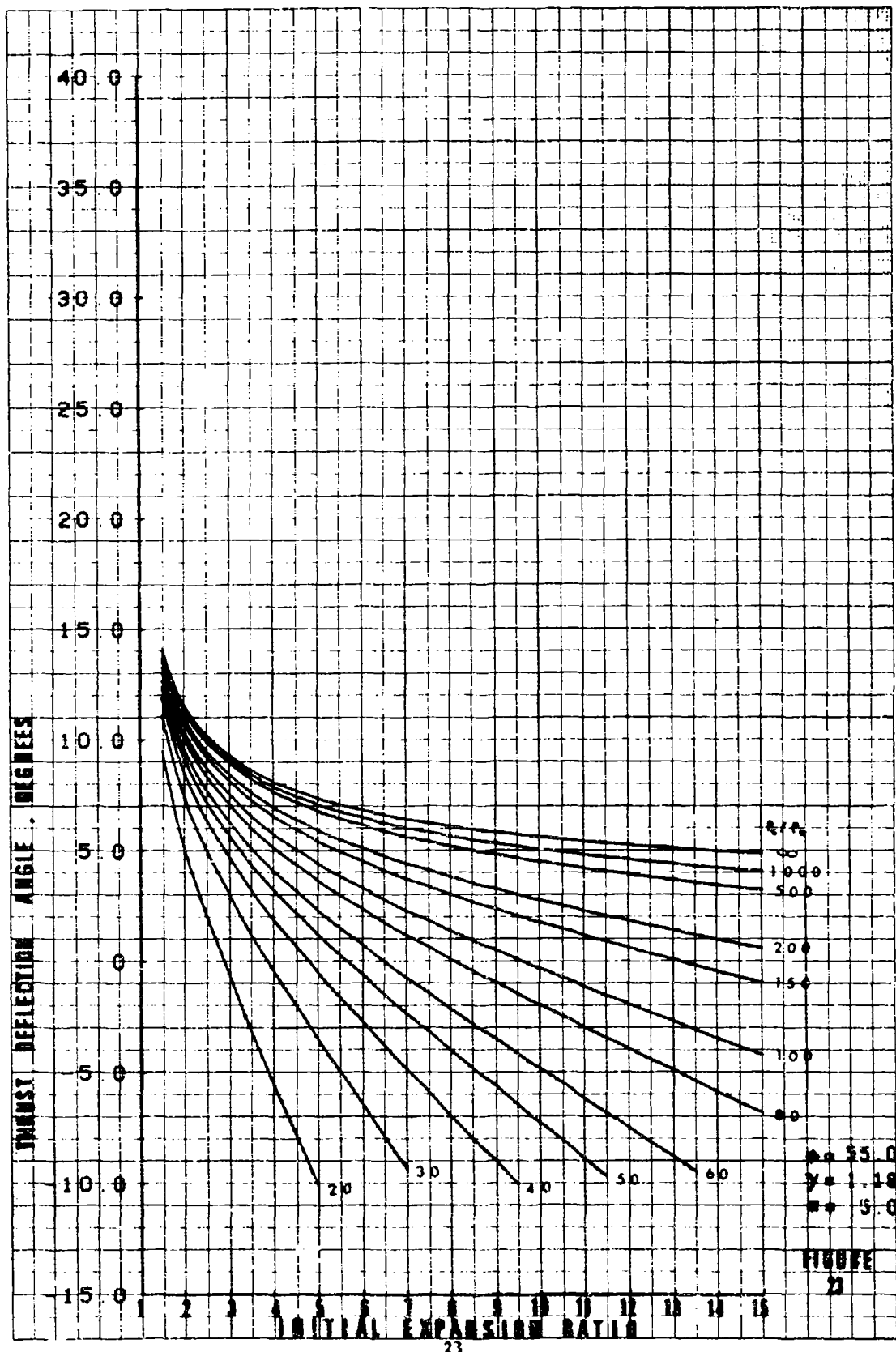


FIGURE 23

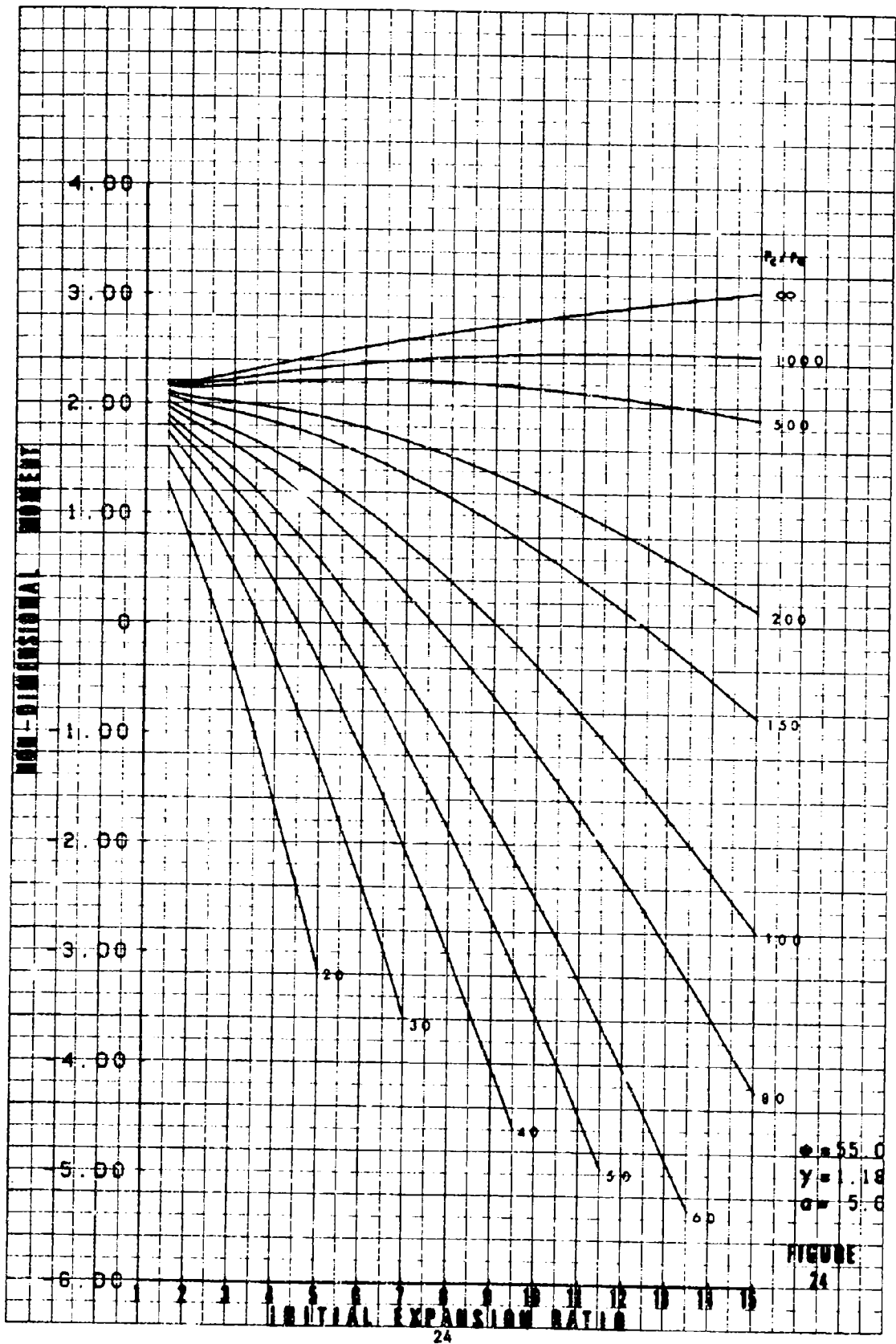
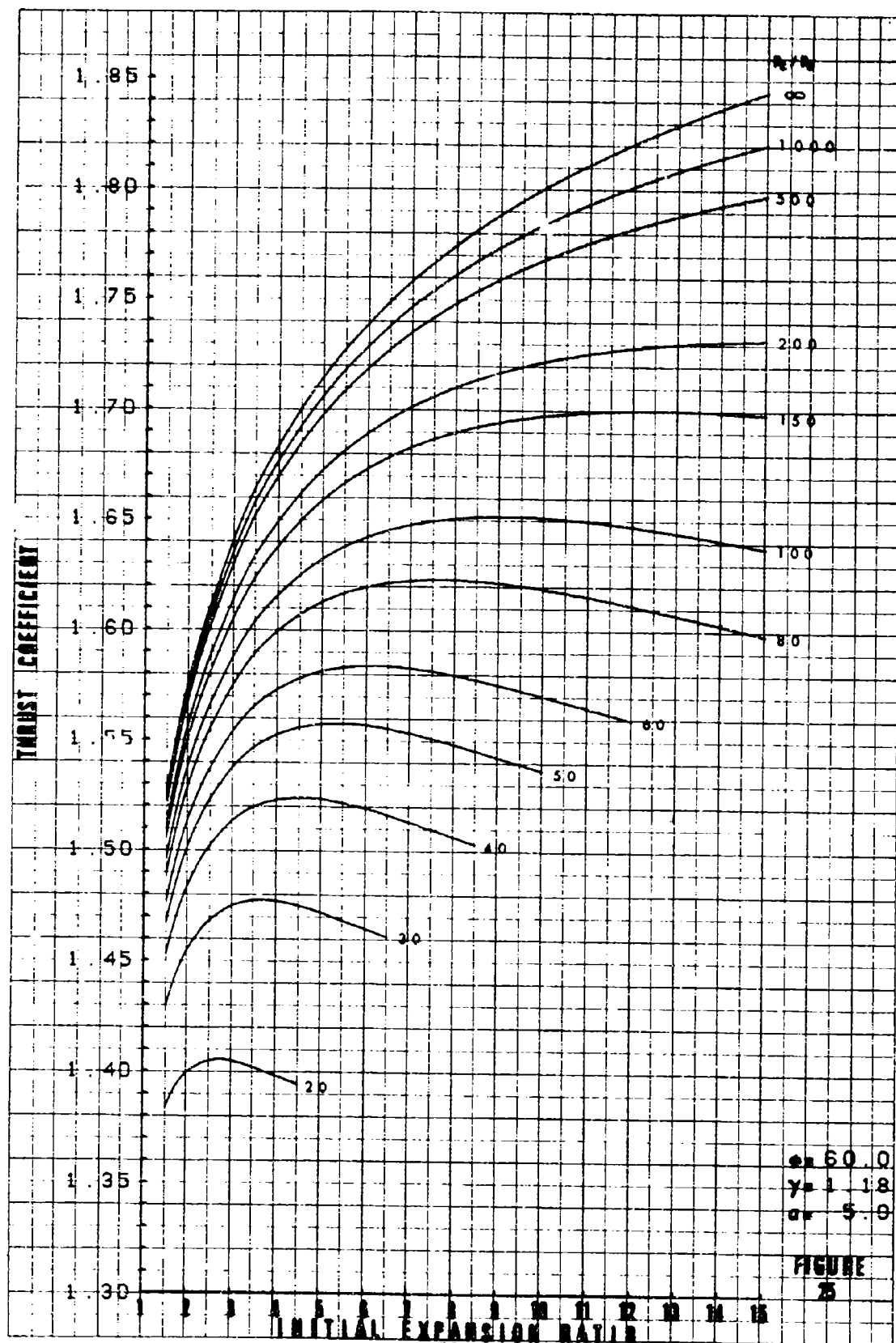
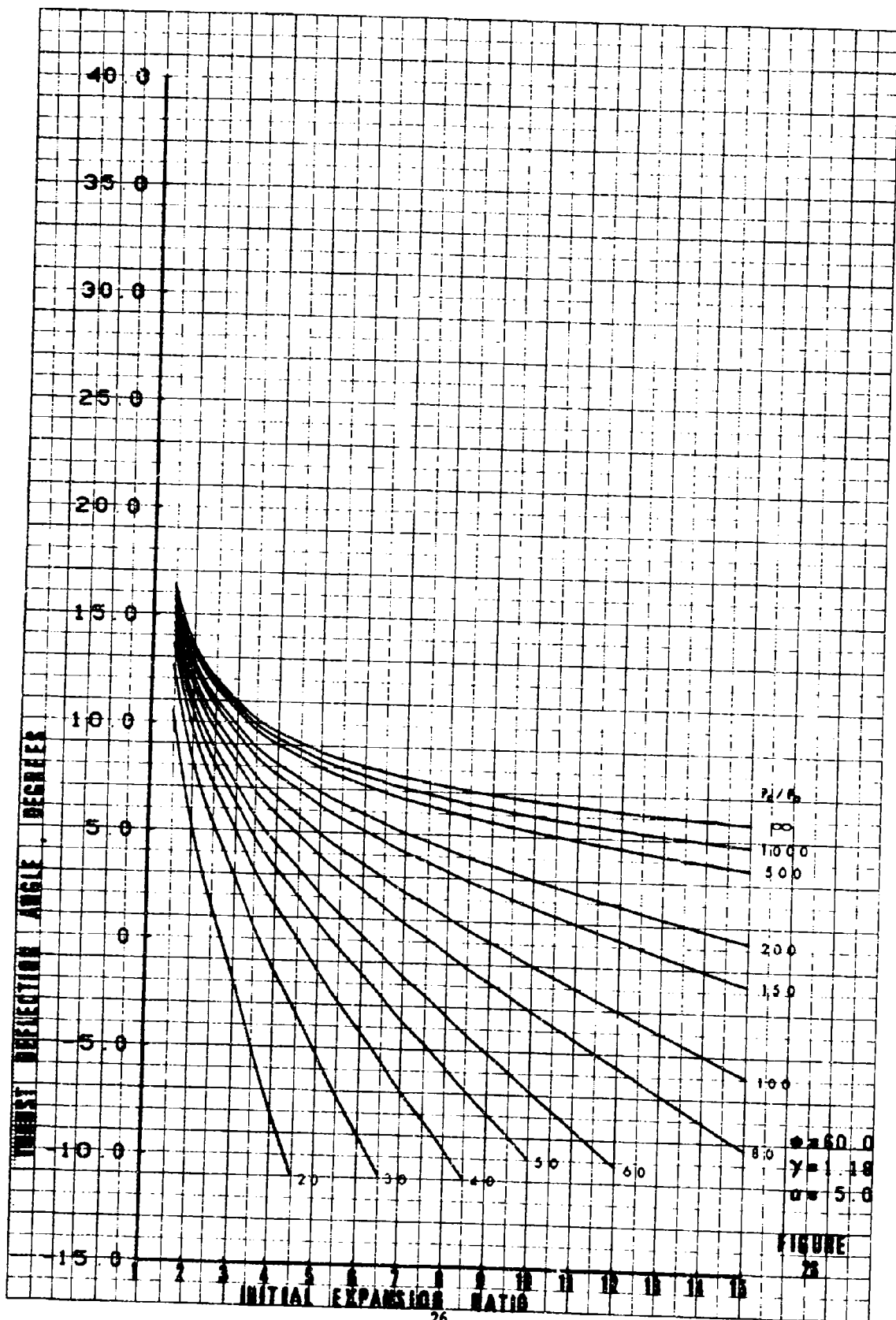
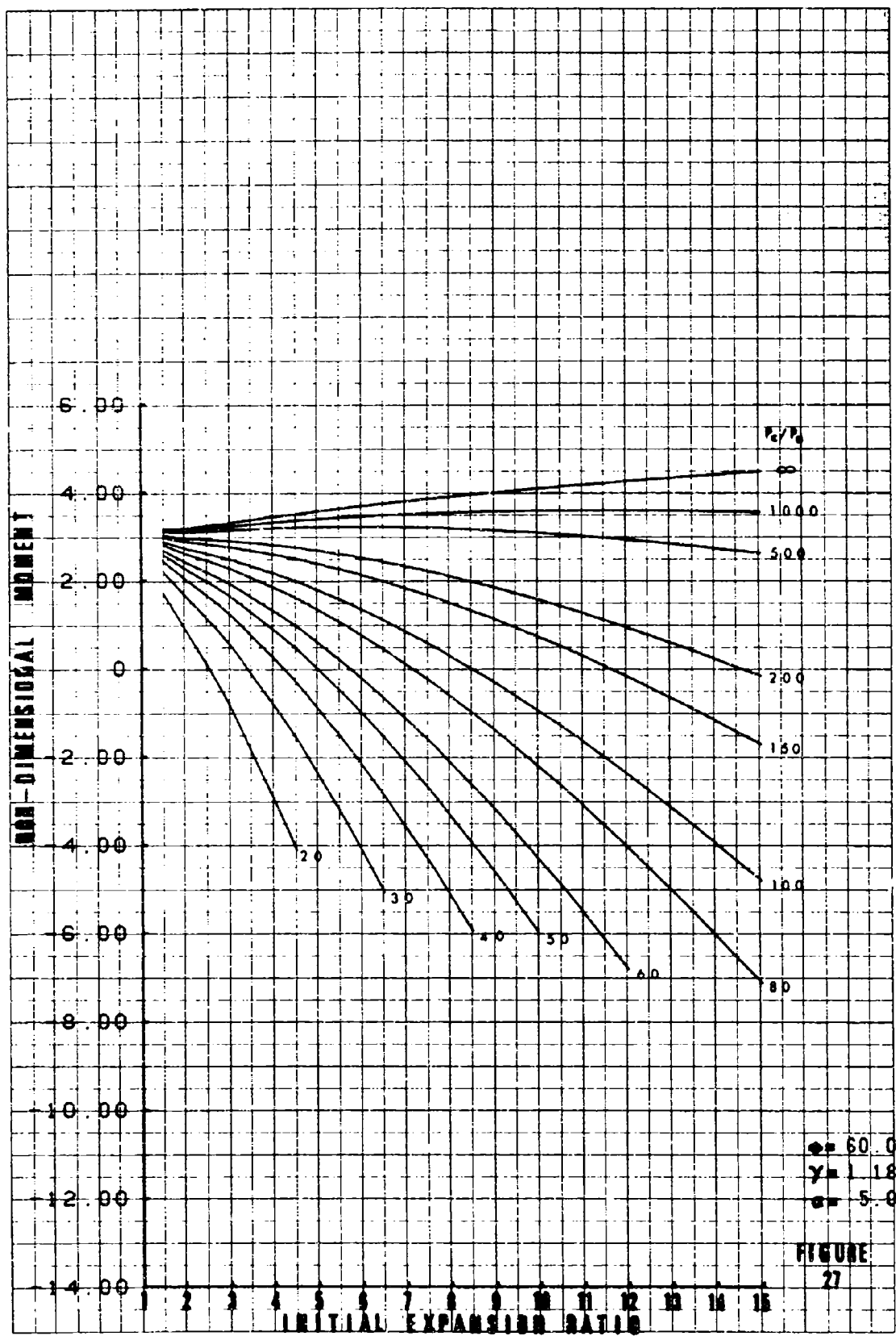
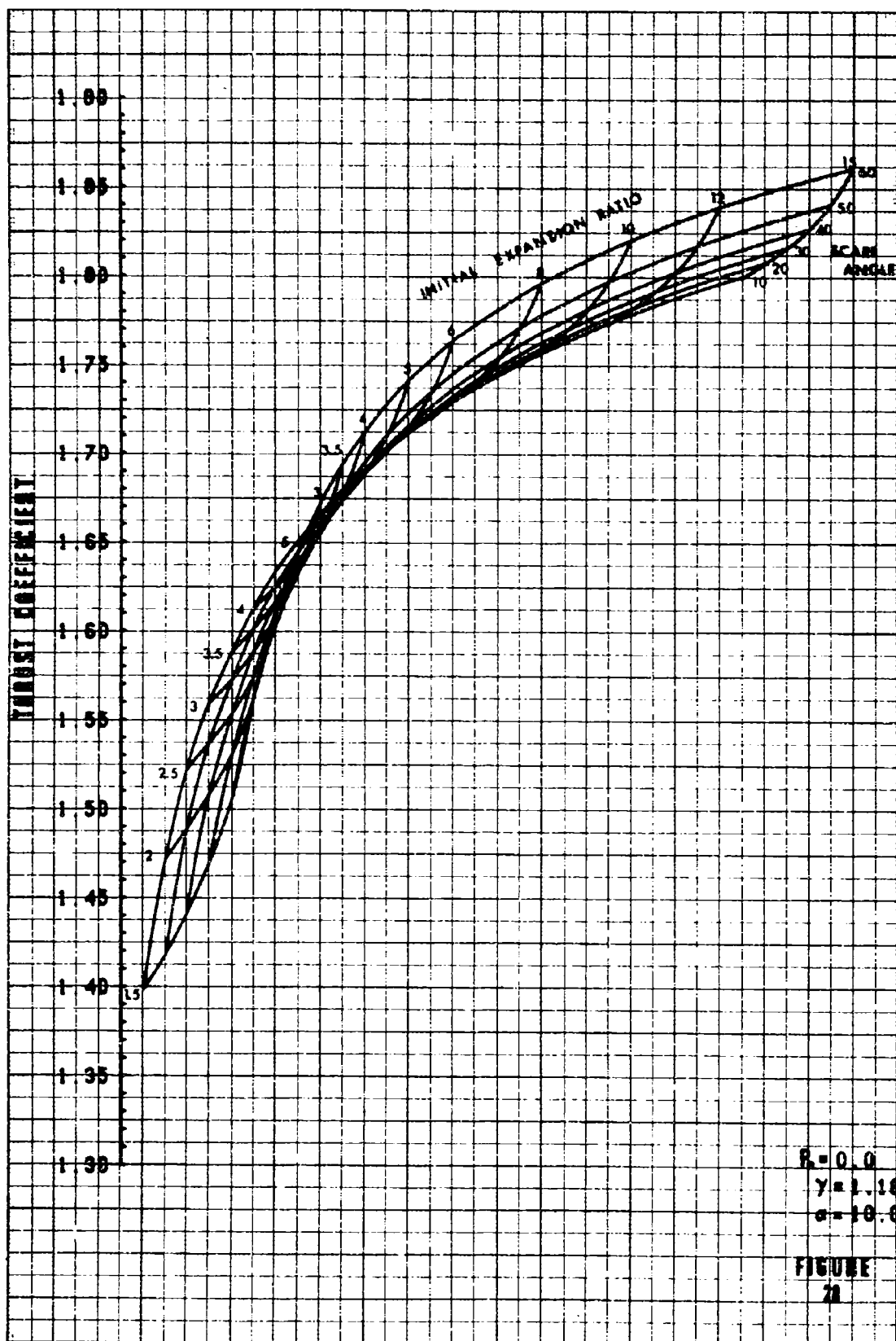


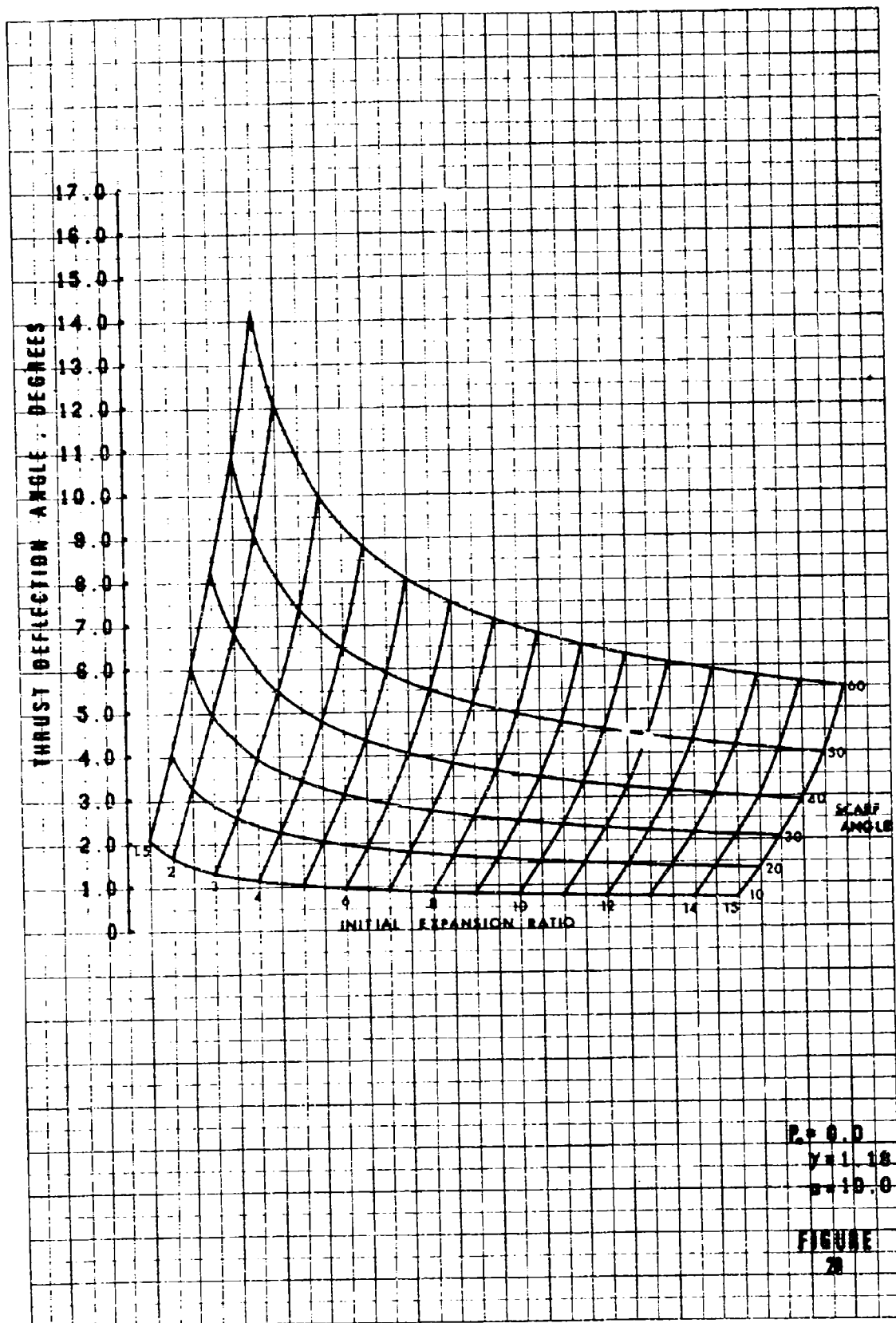
FIGURE 24

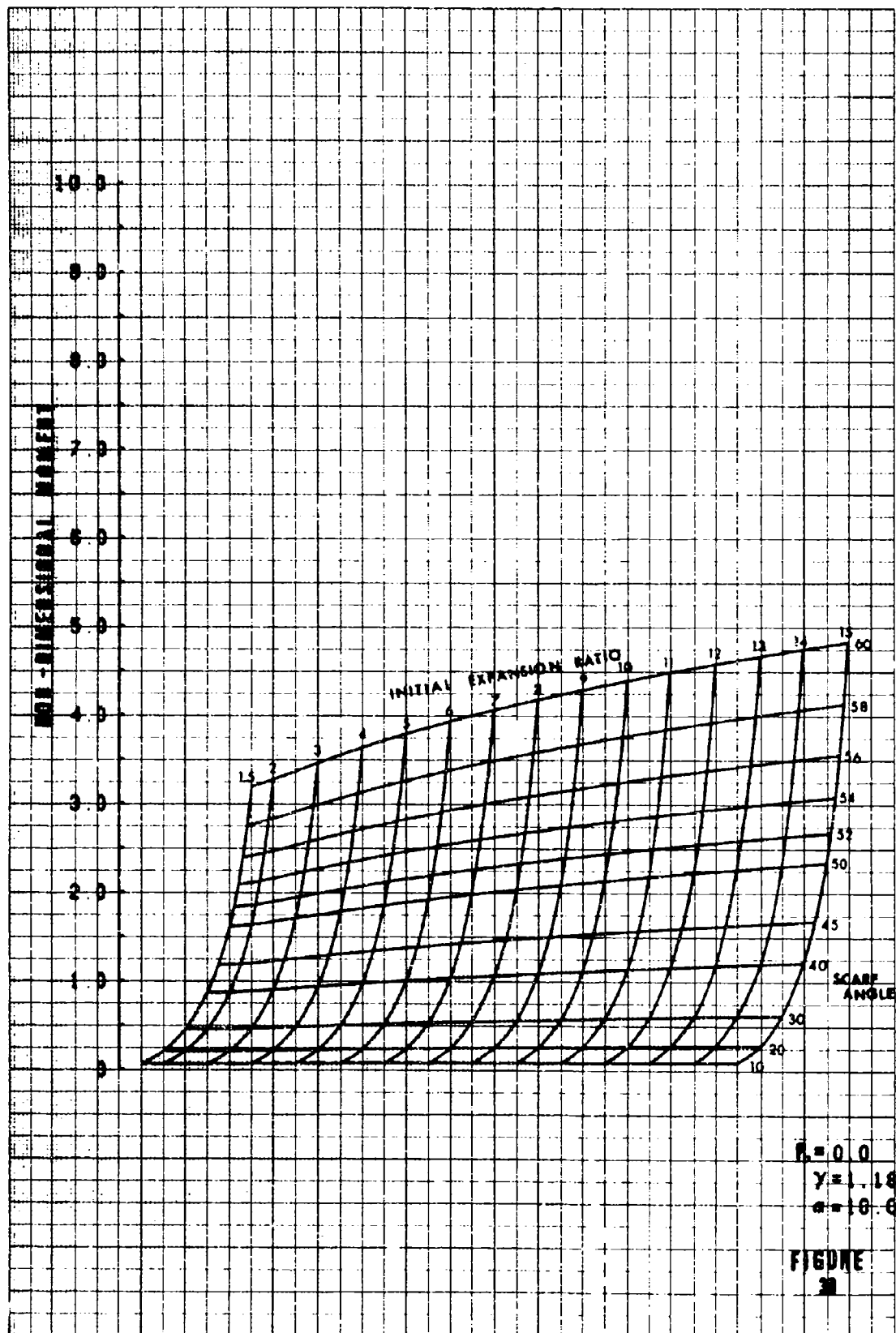


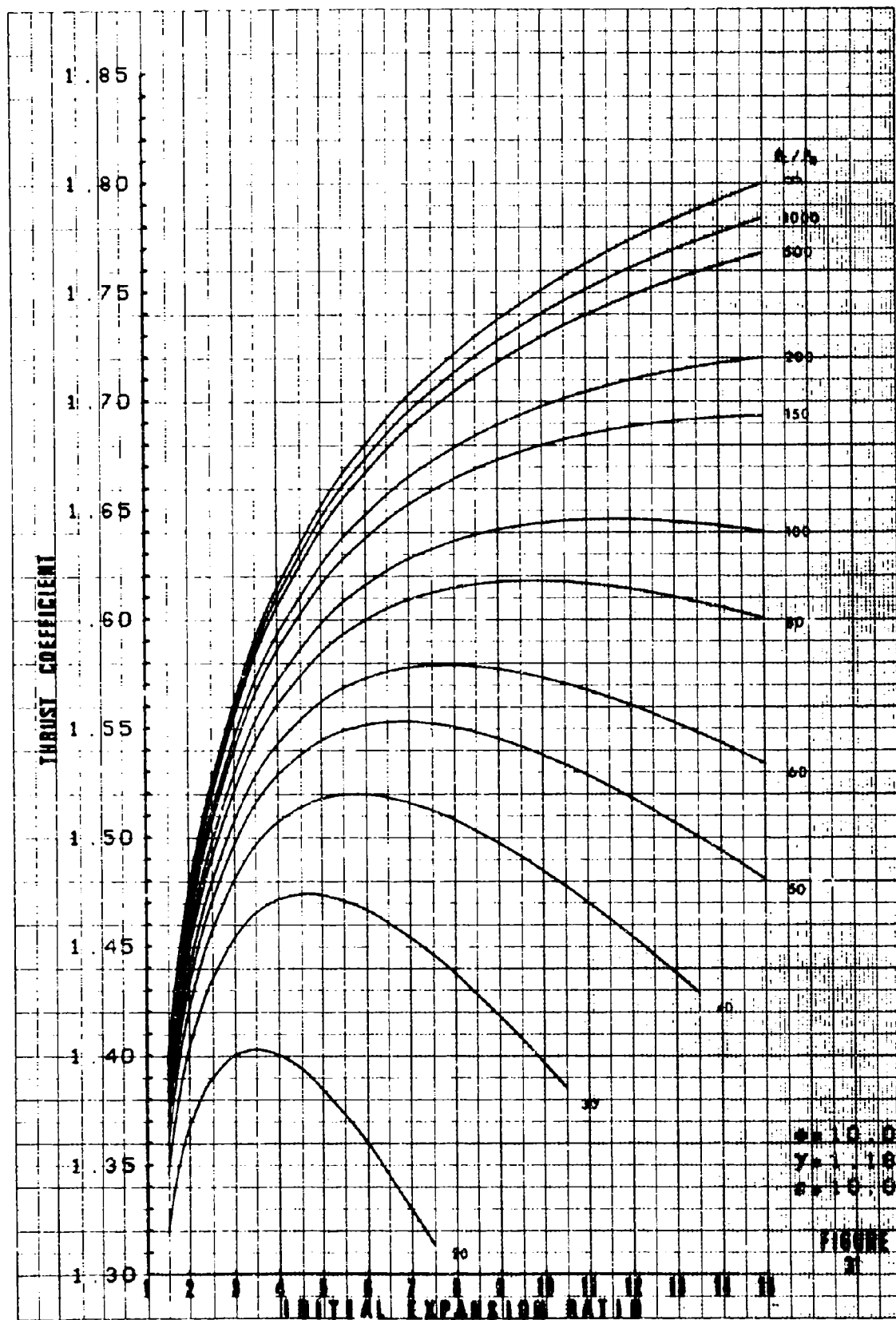


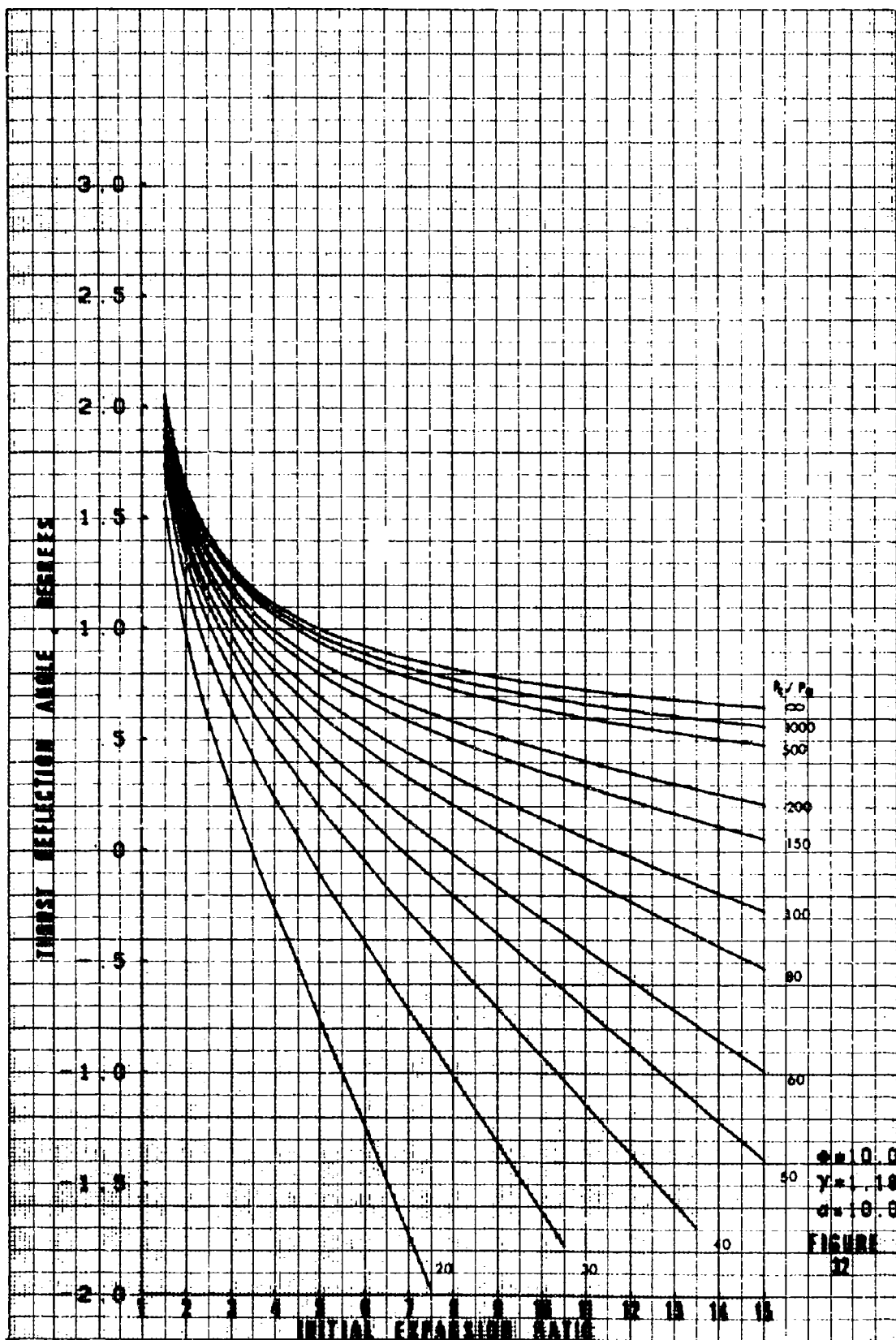




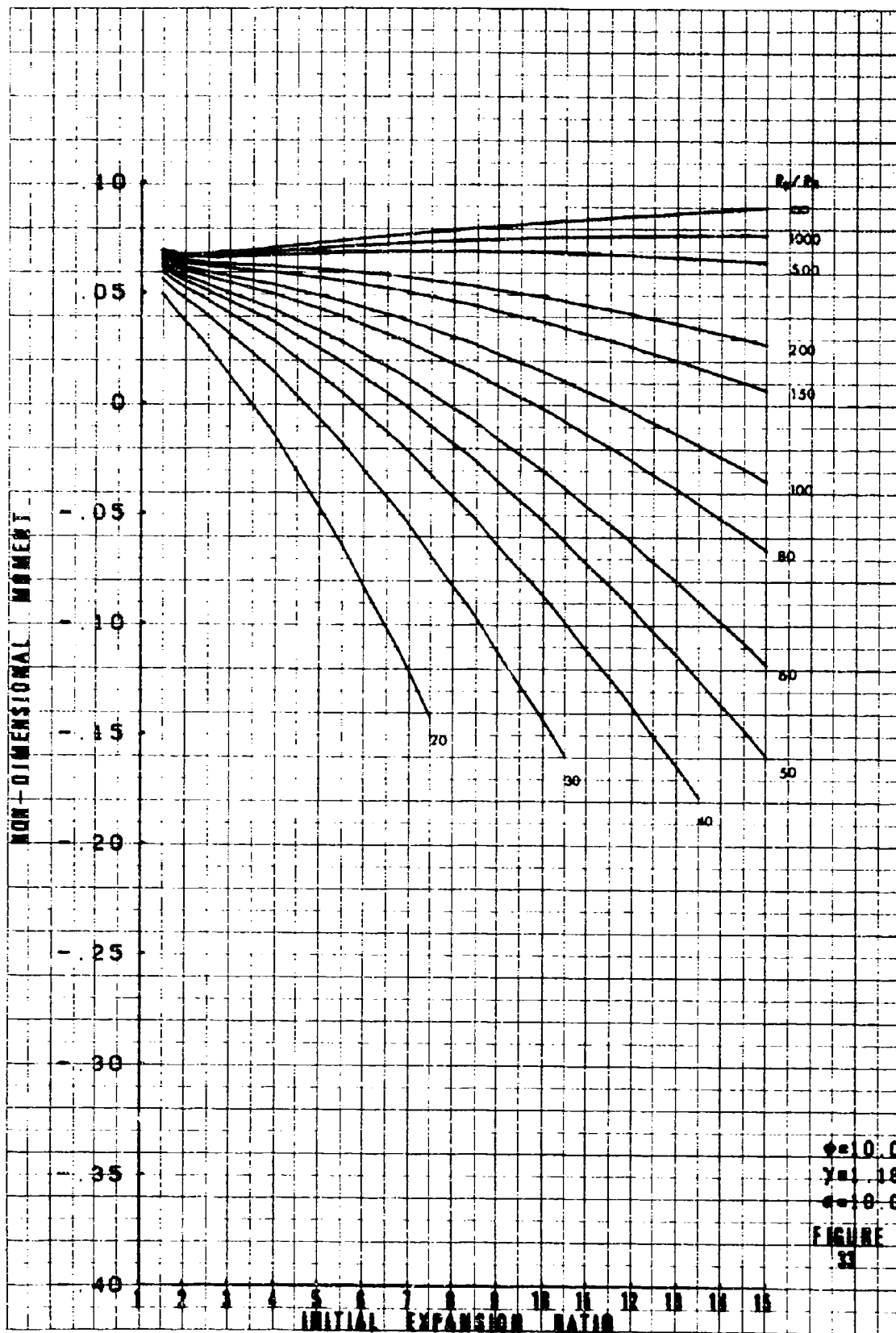


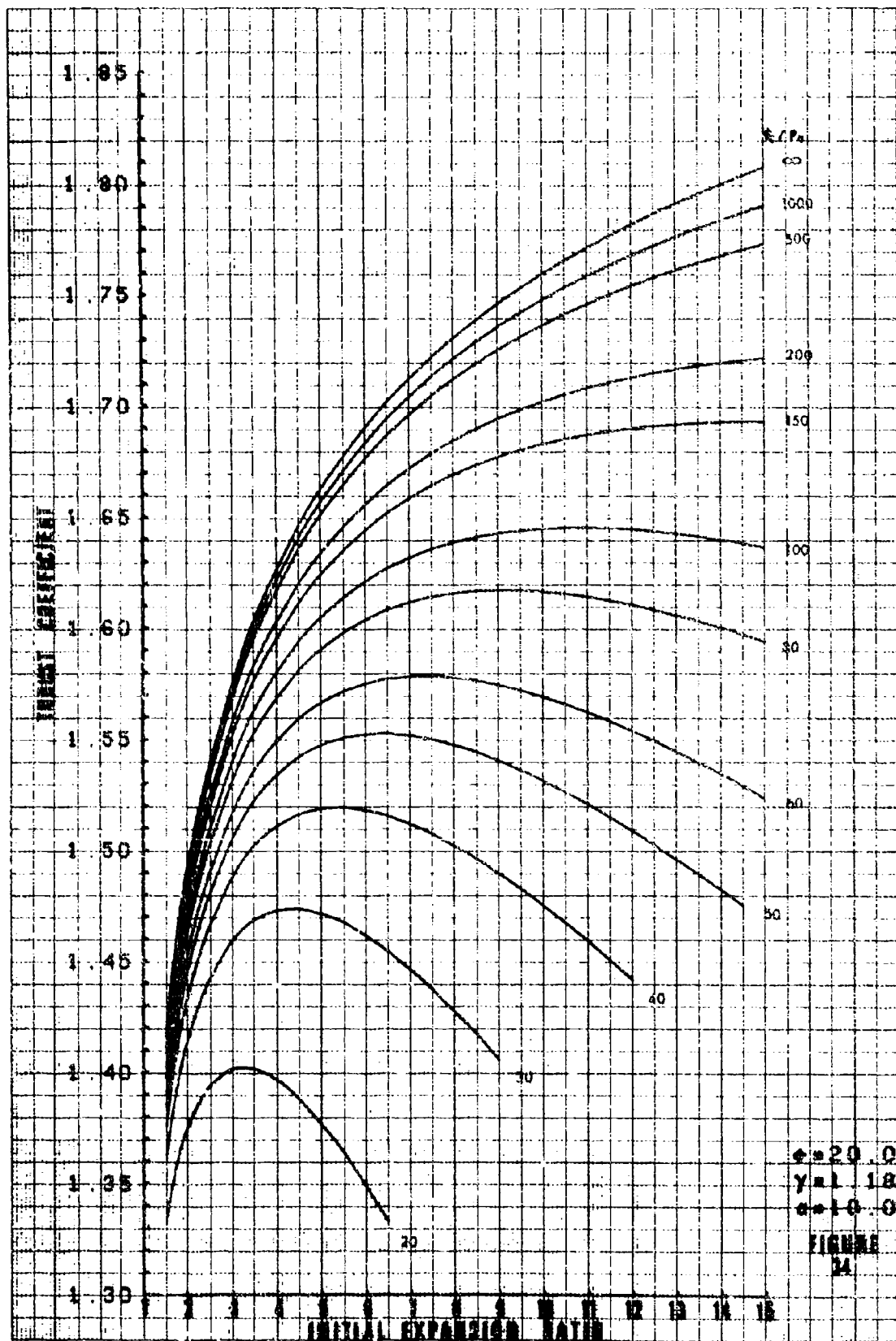


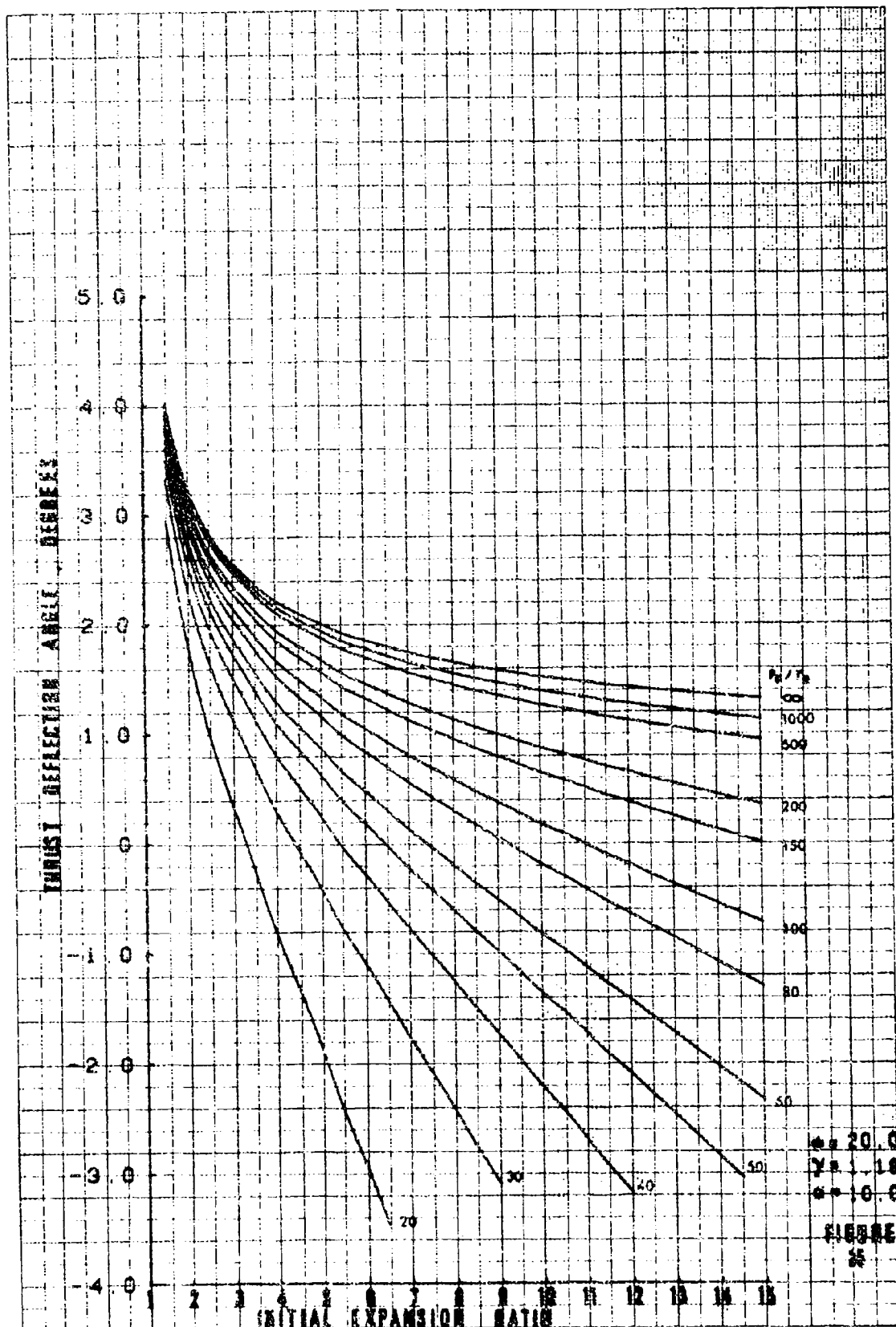


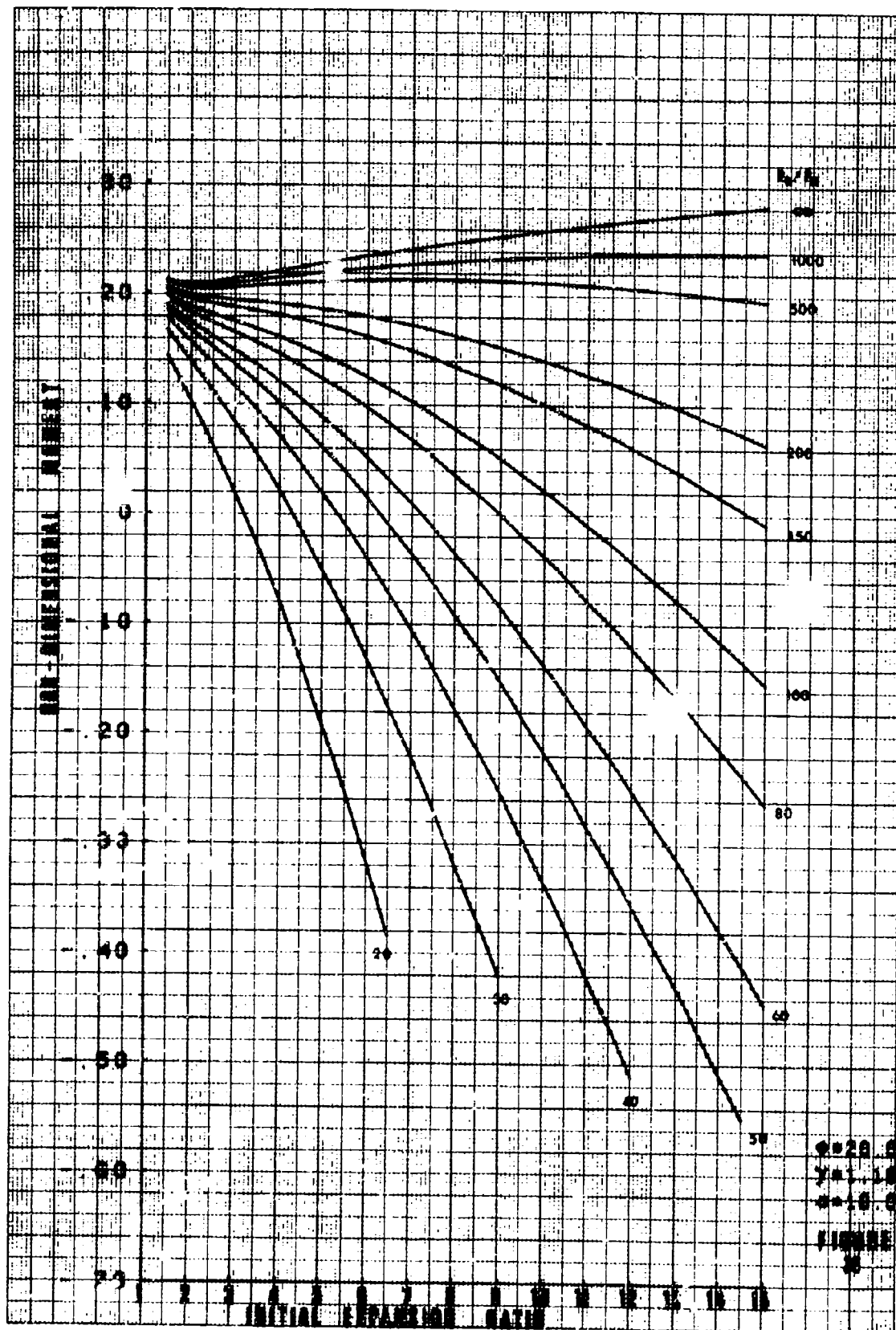


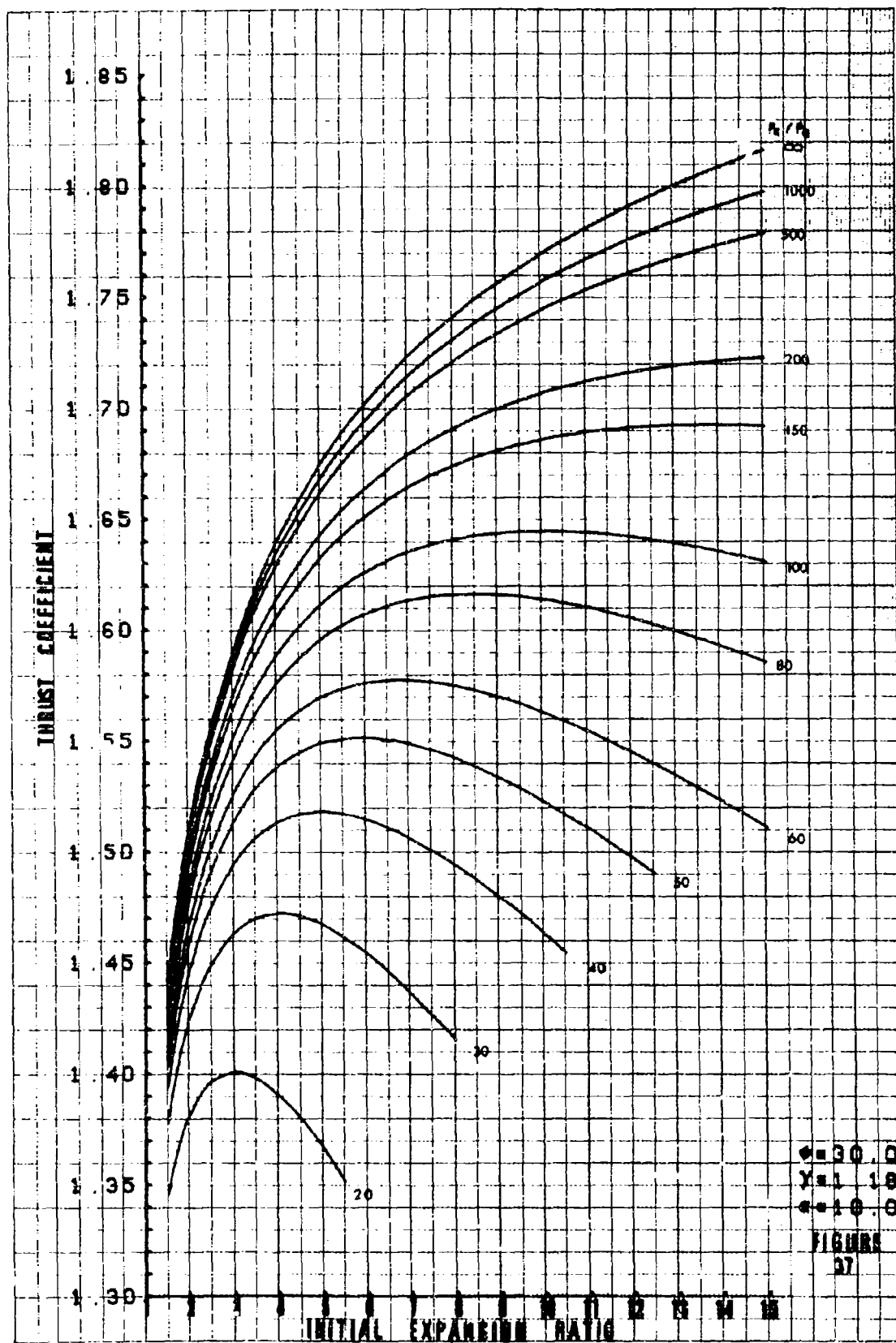
$\gamma = 1.4$
 $\alpha = 10^\circ$
FIGURE 32











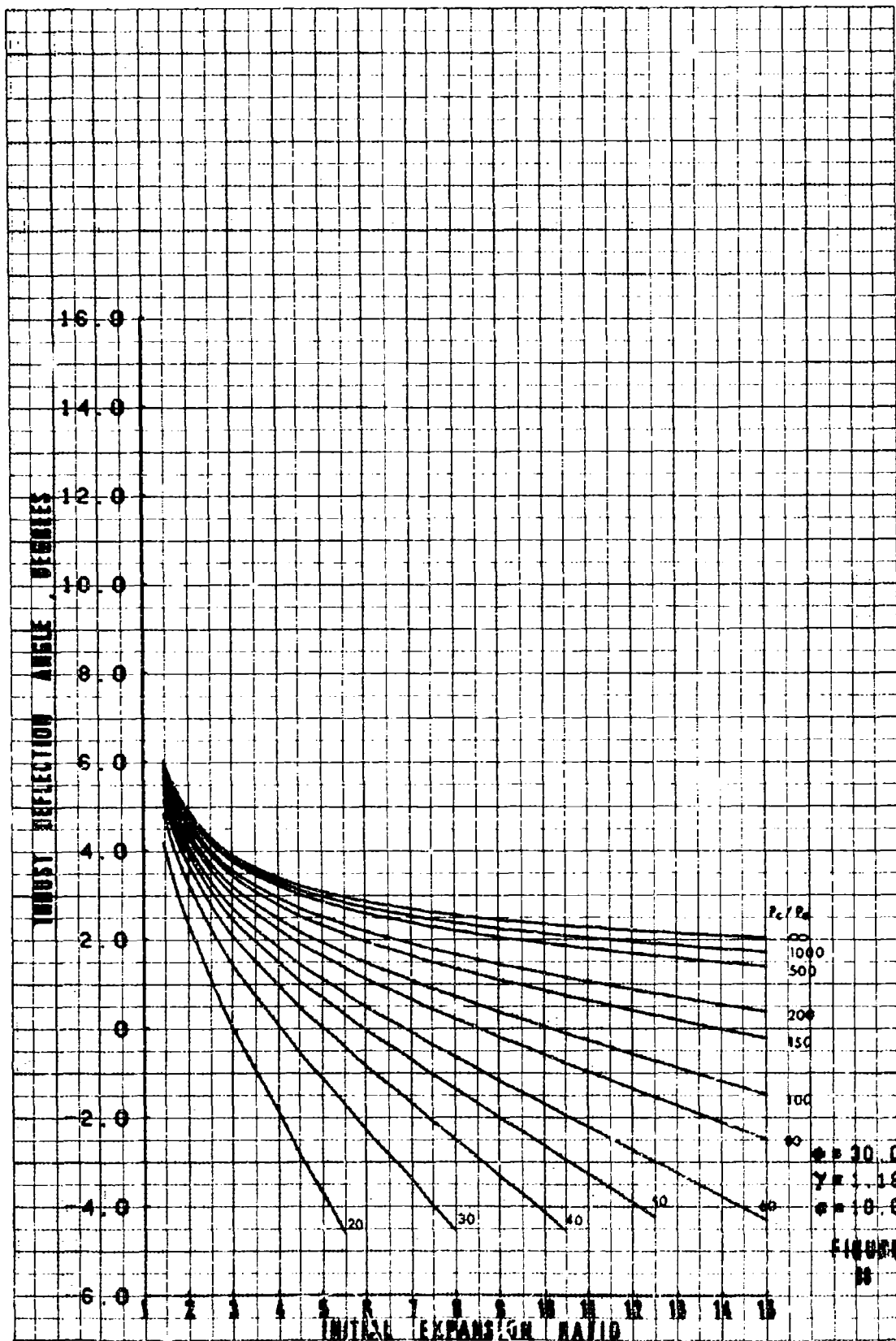
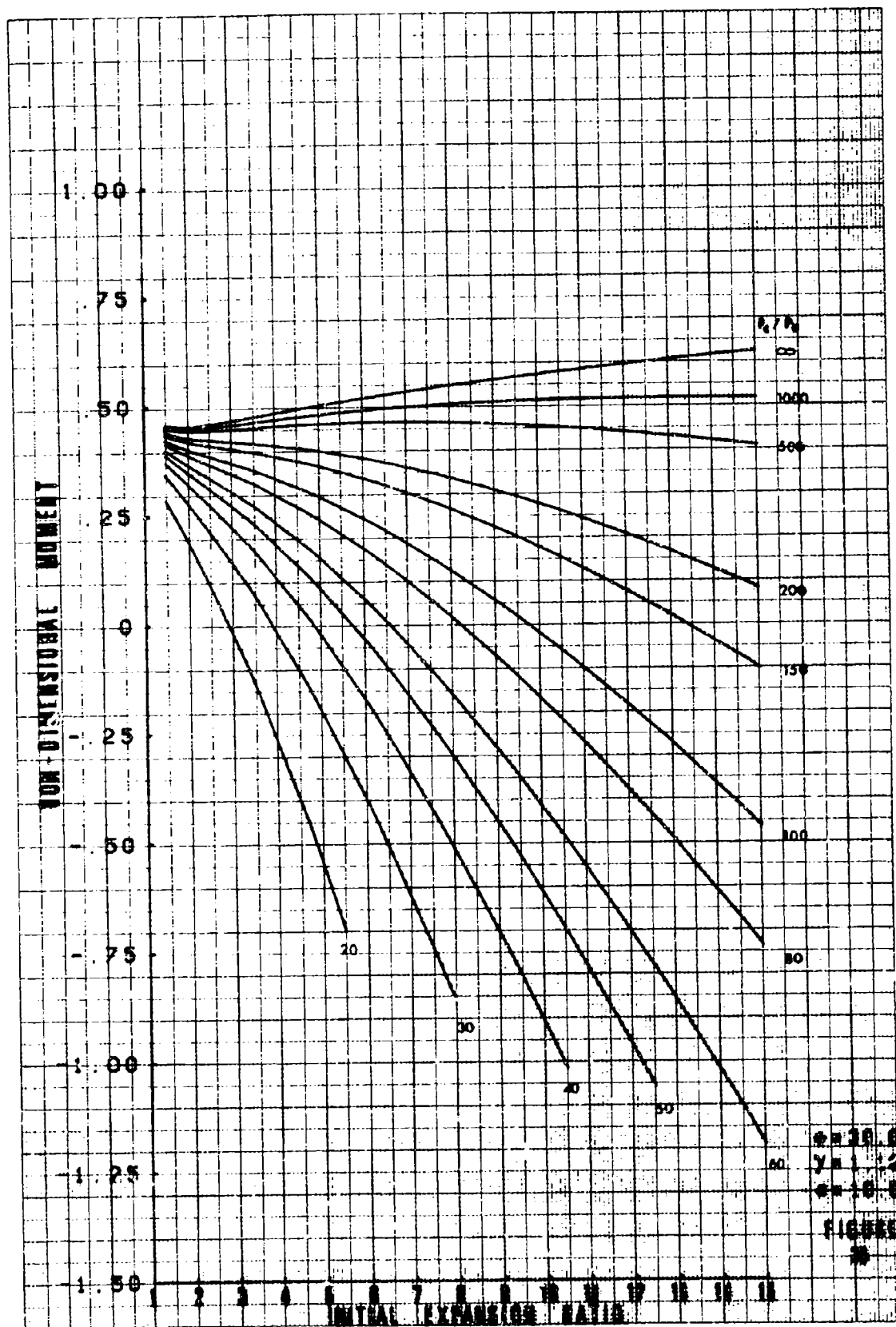
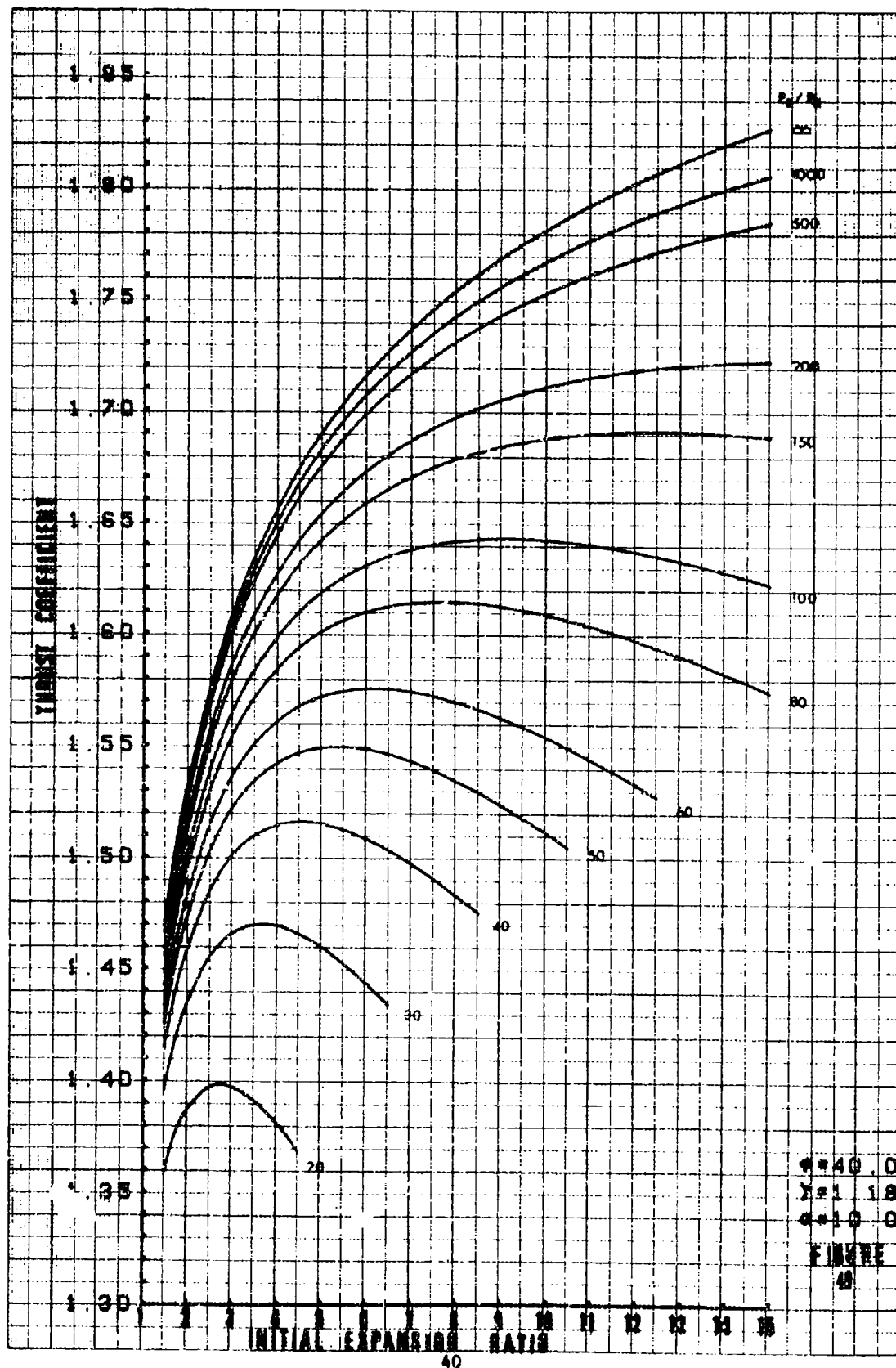
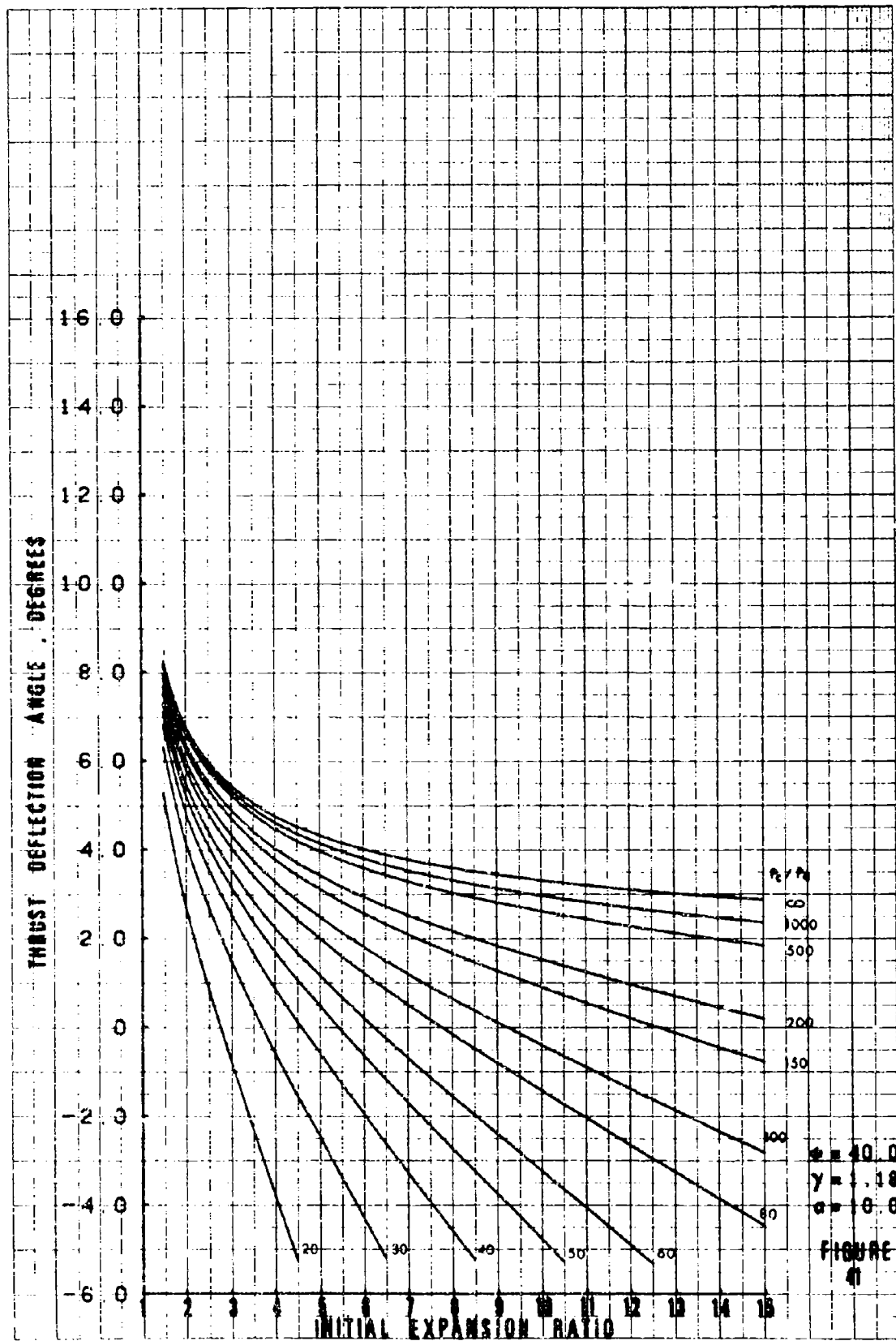
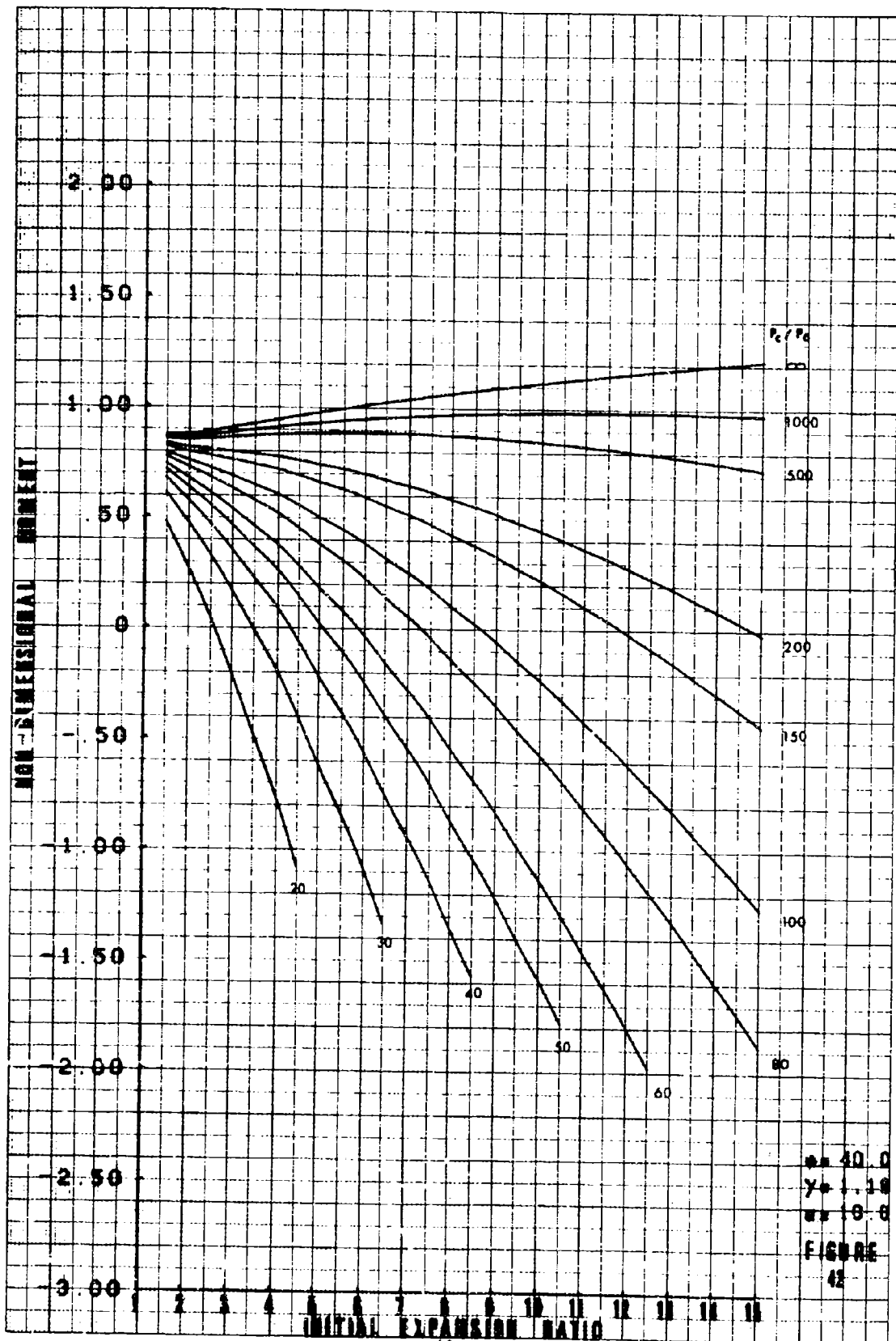


FIGURE 5









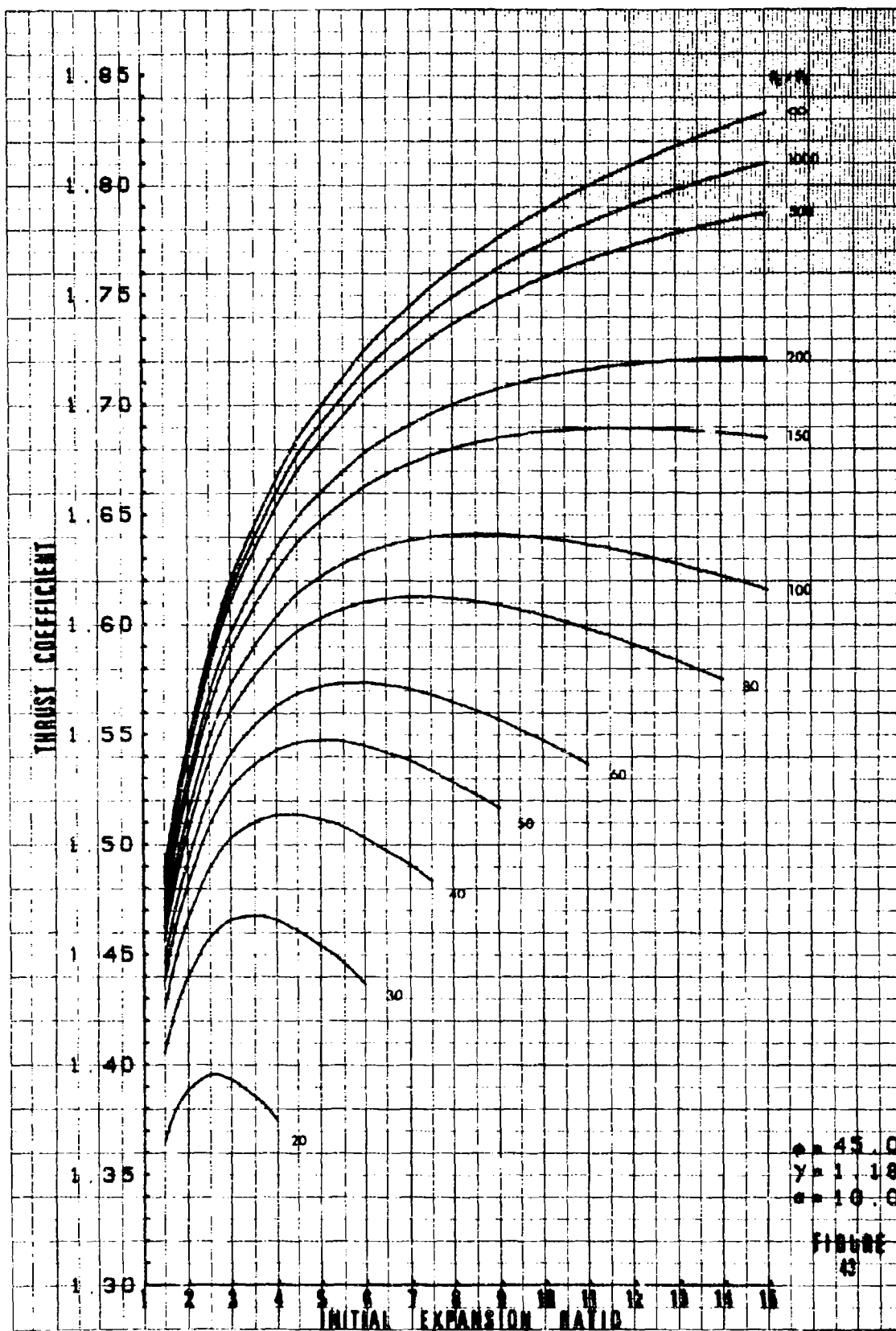
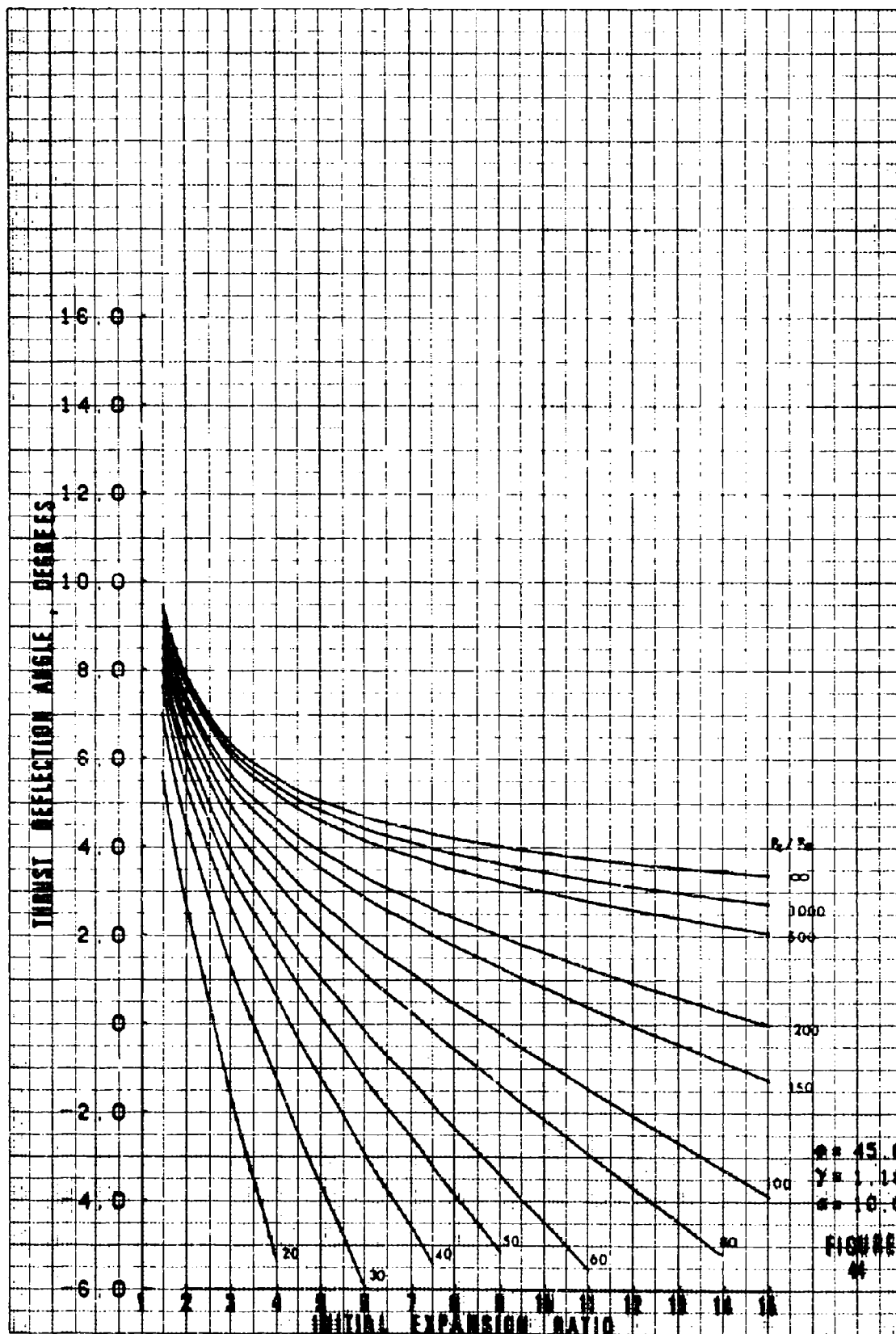
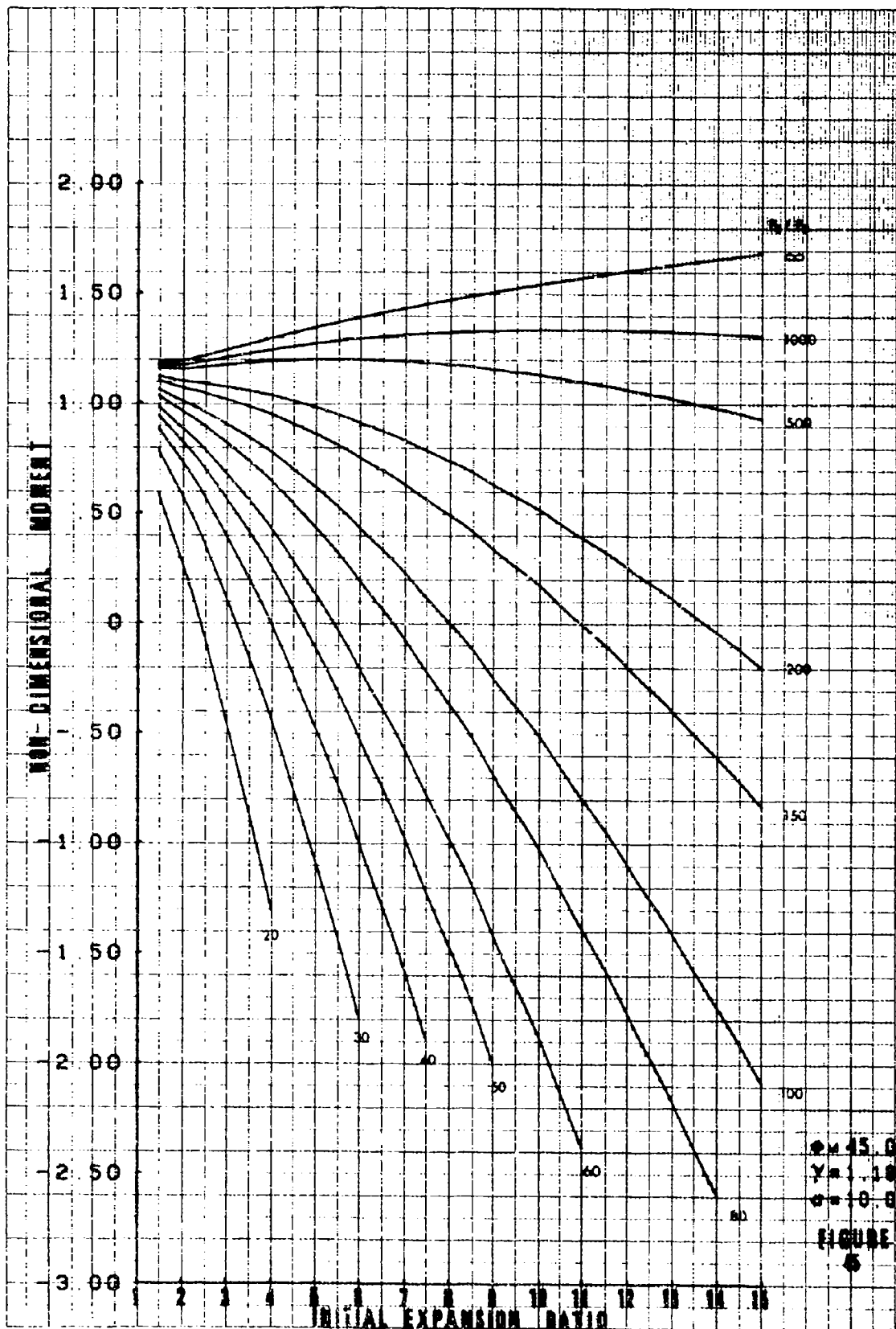


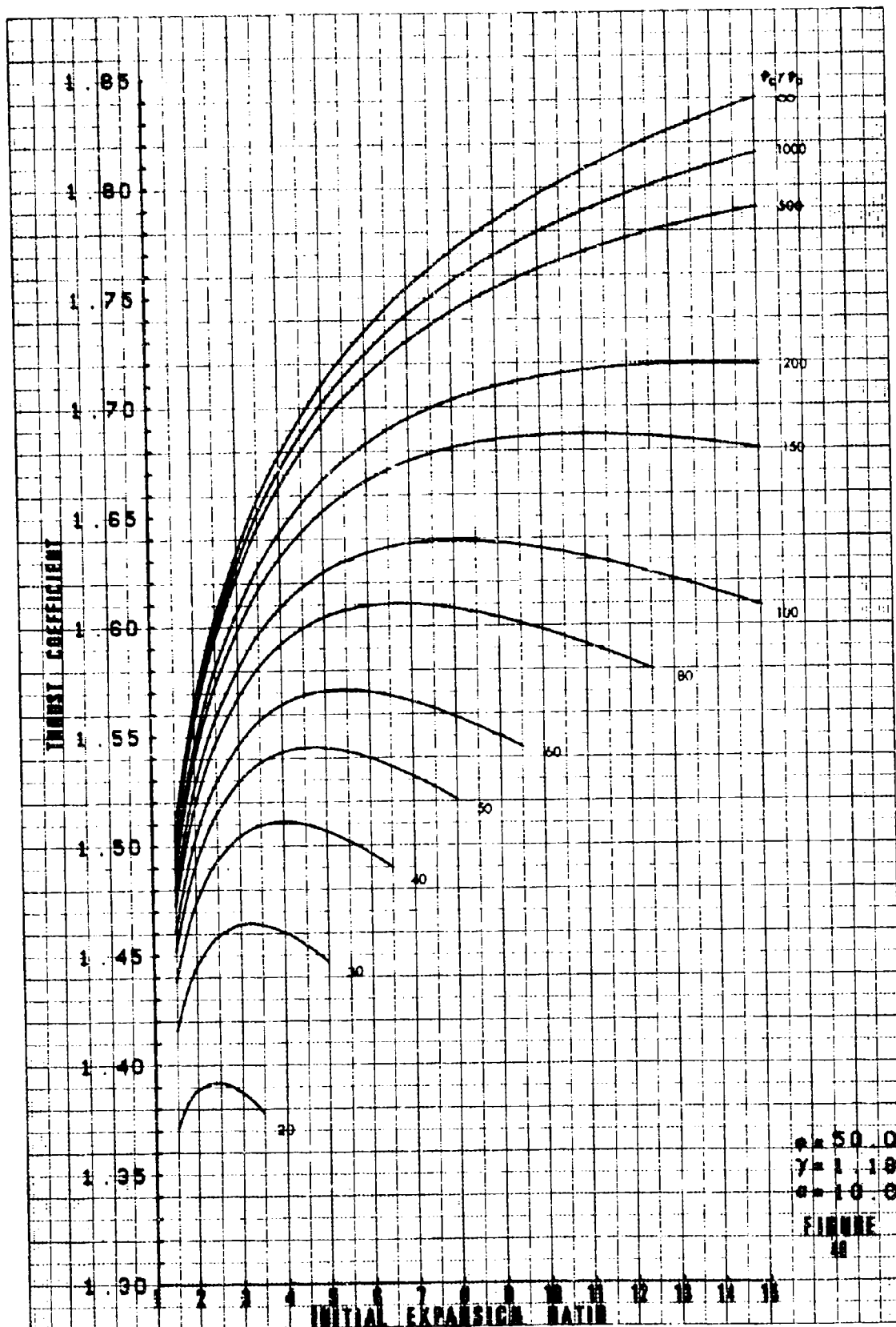
FIGURE 43

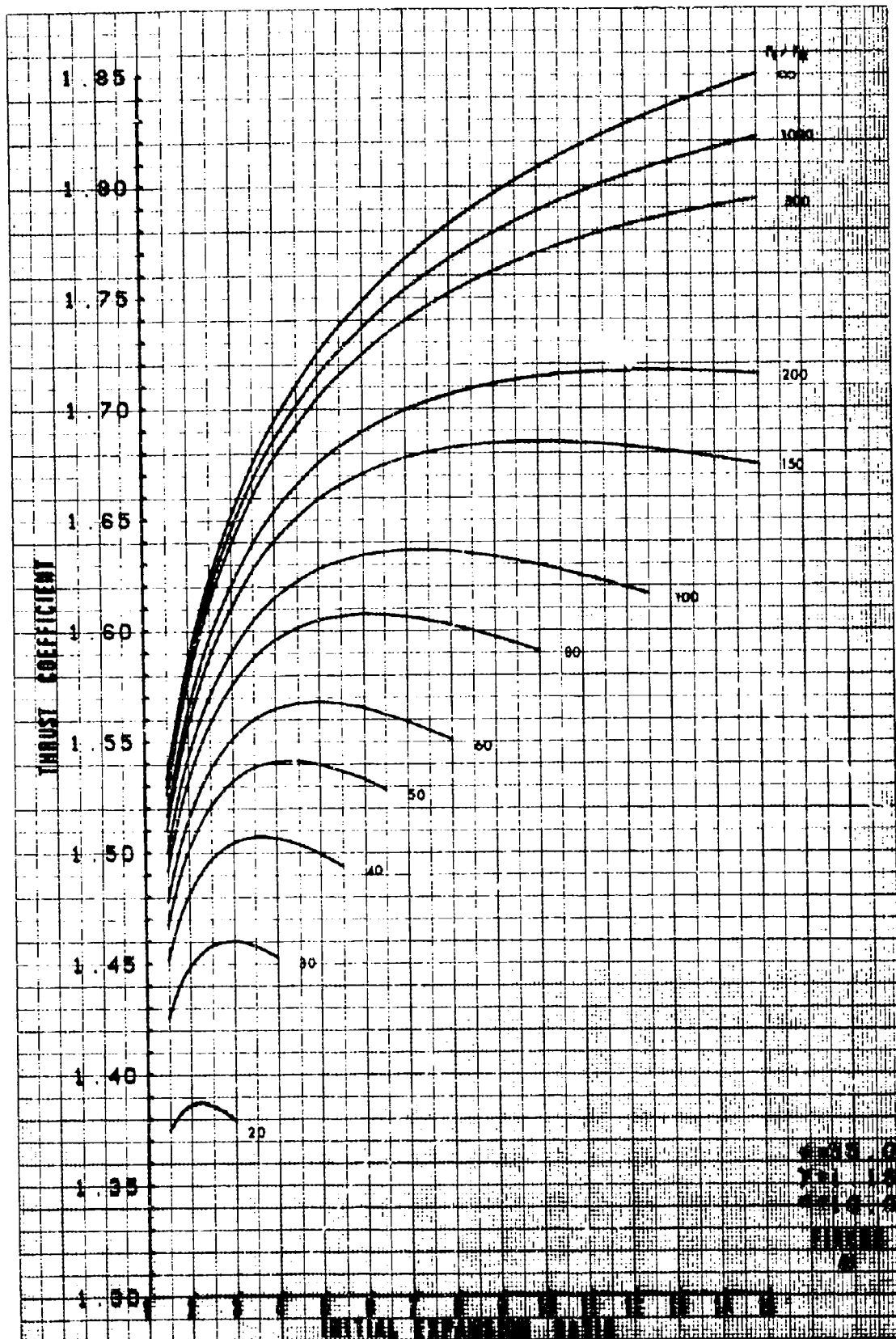


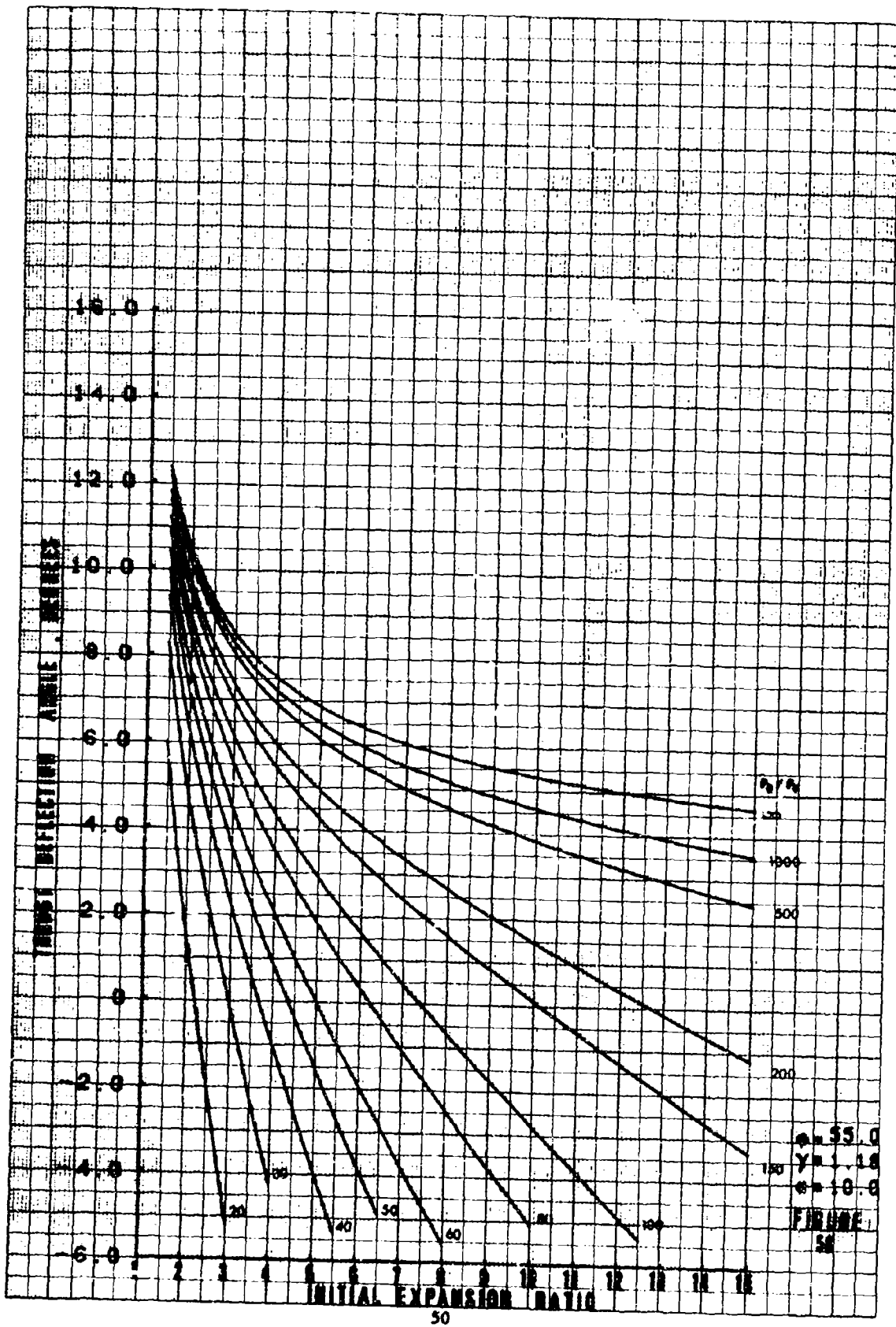
$\theta = 45.0$
 $\gamma = 1.4$
 $\sigma = 10.0$

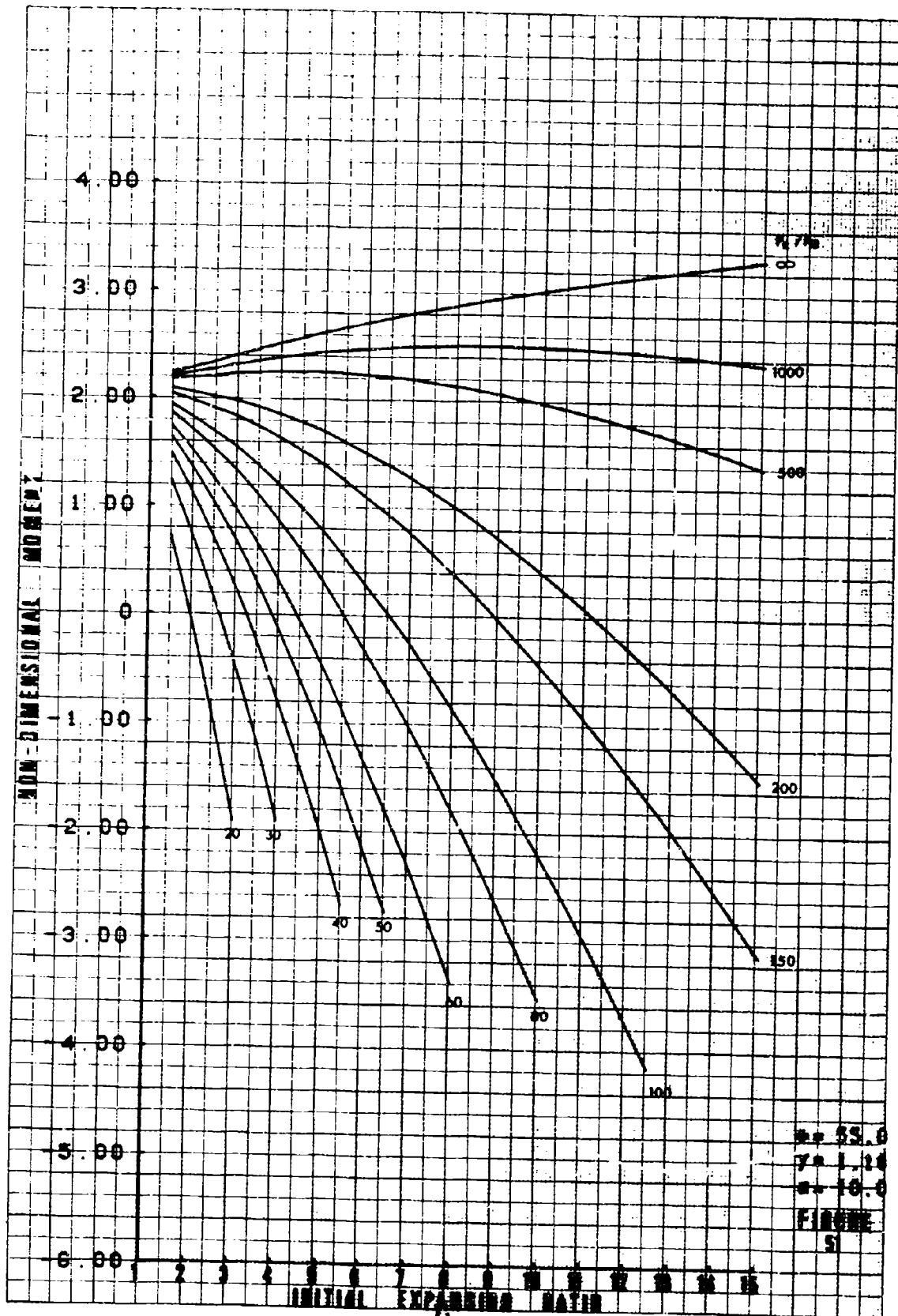
FIGURE 4

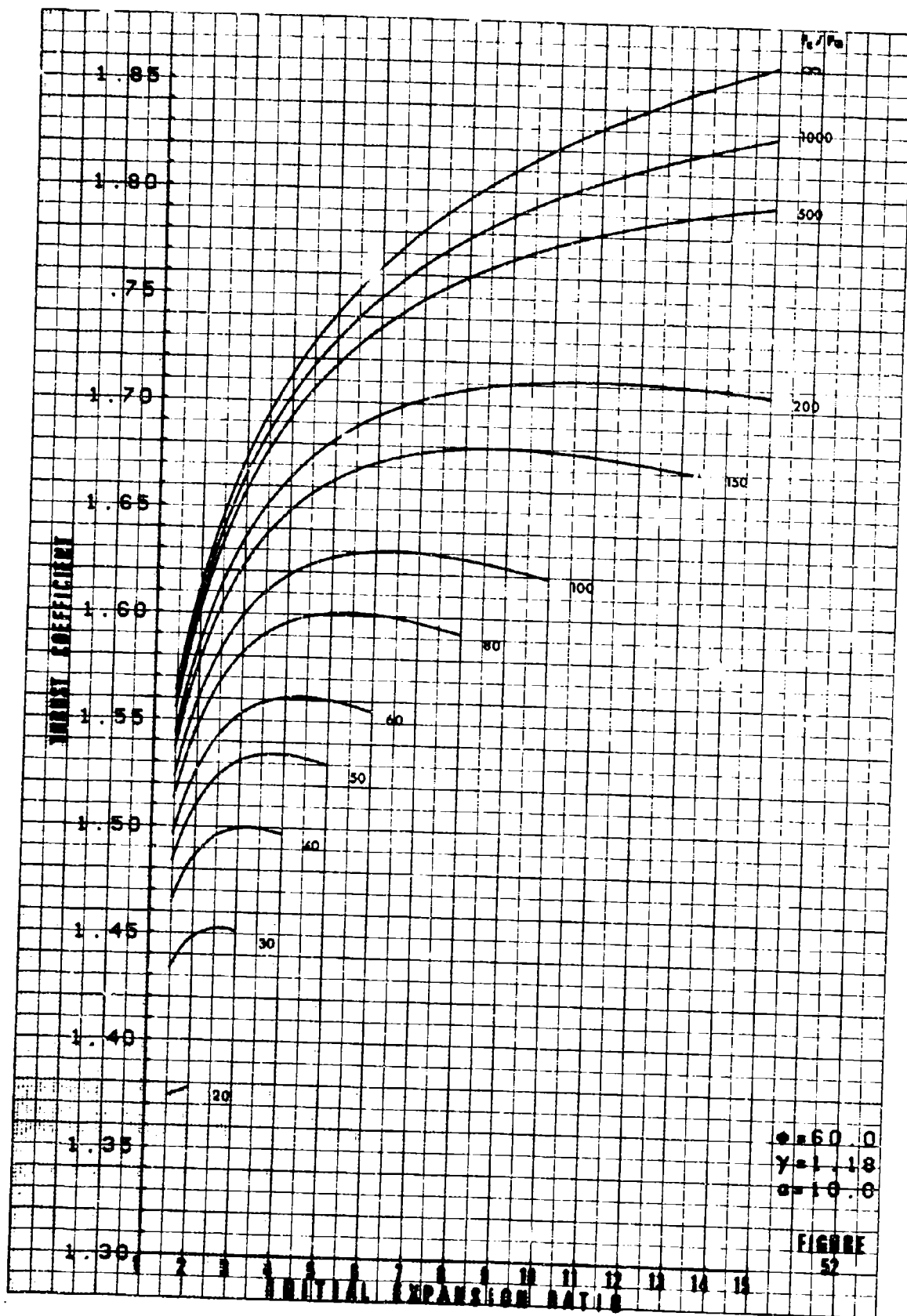


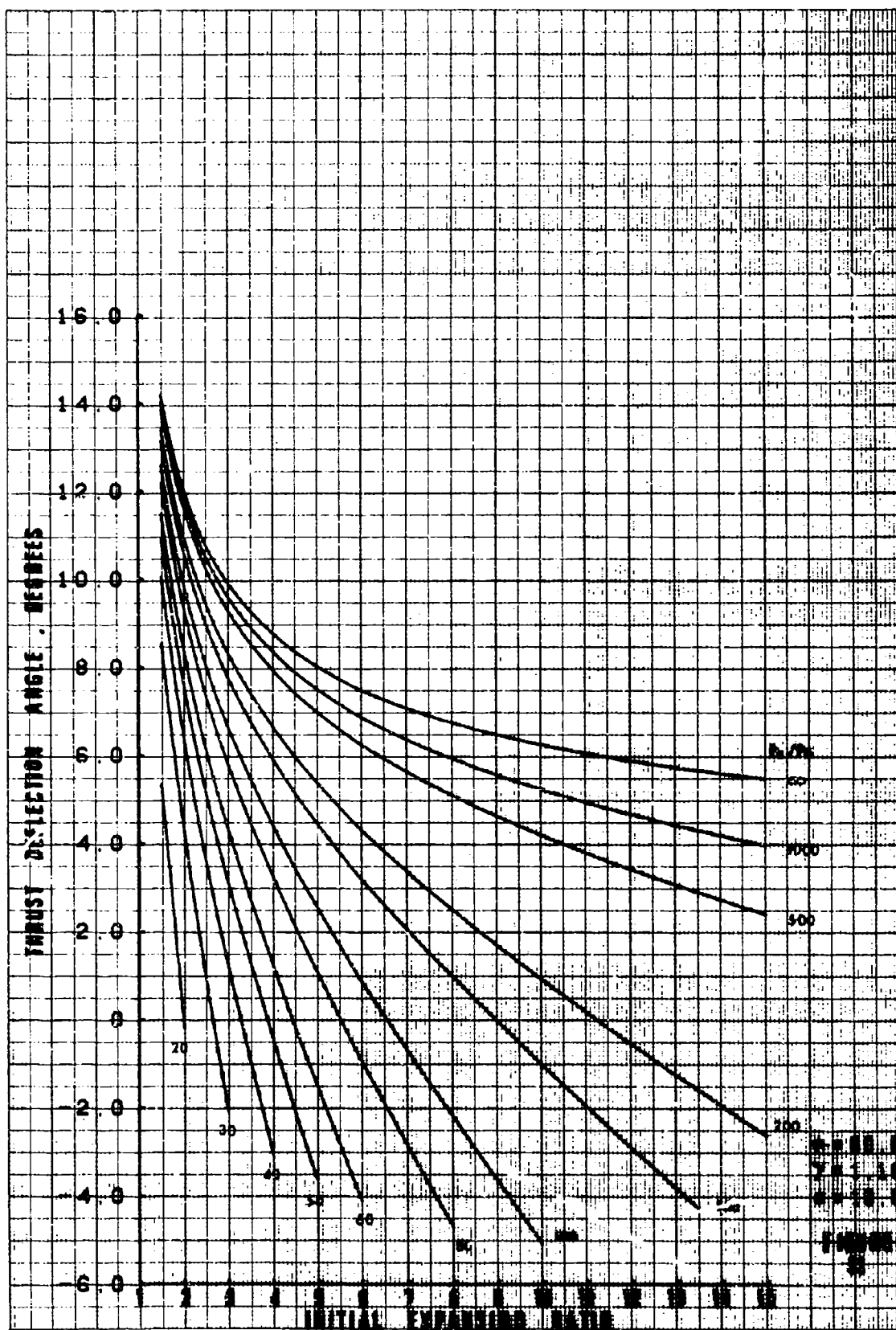


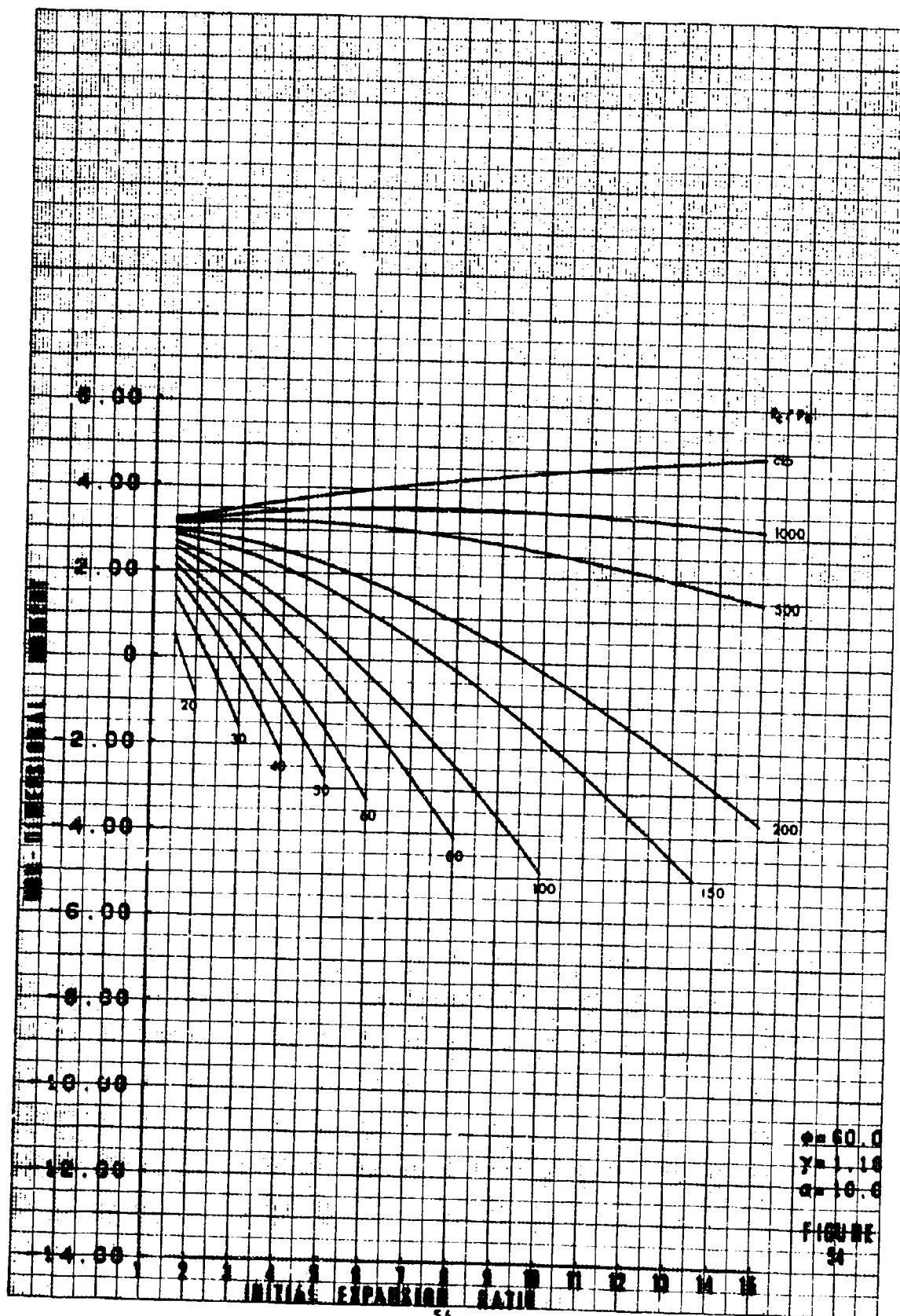


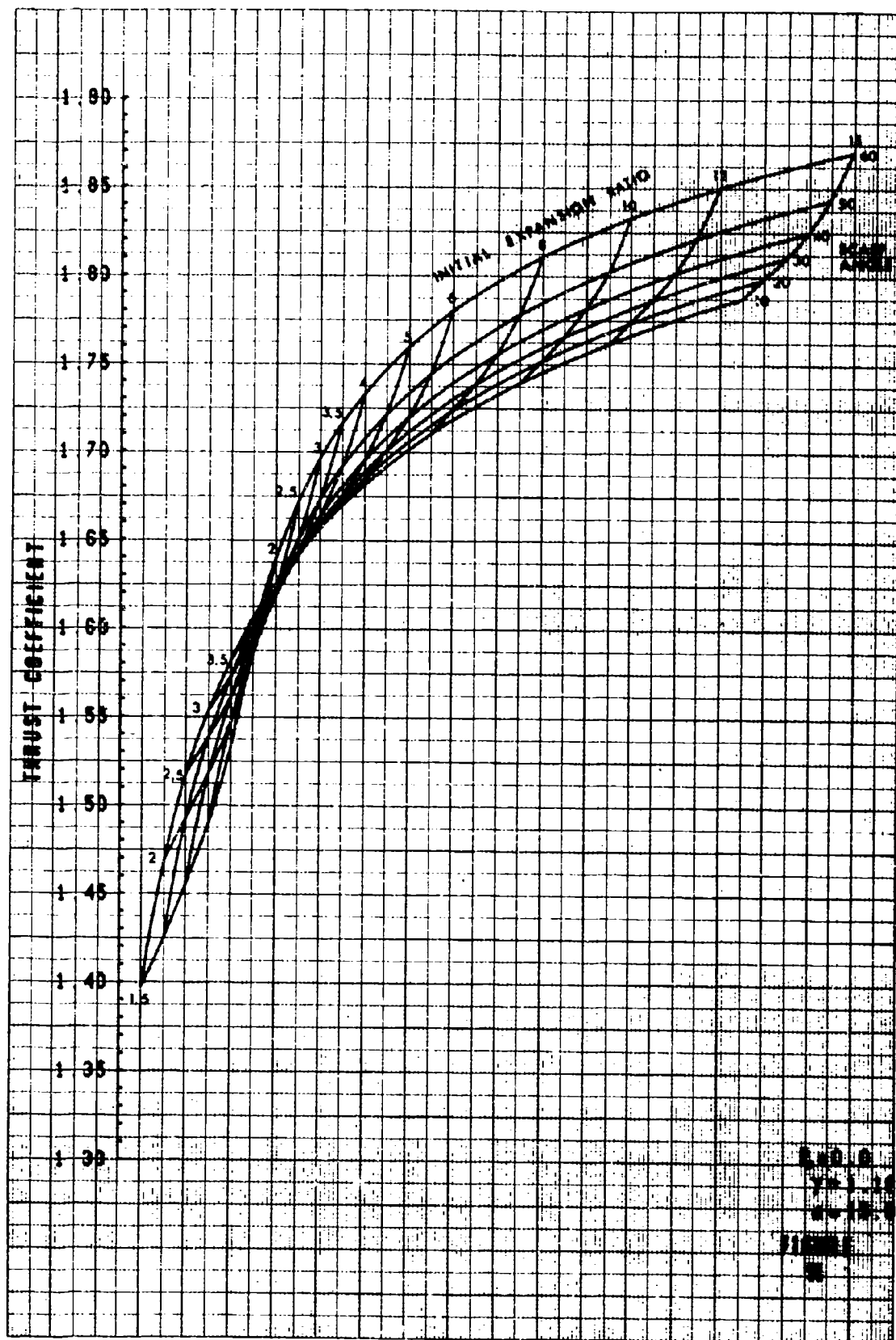


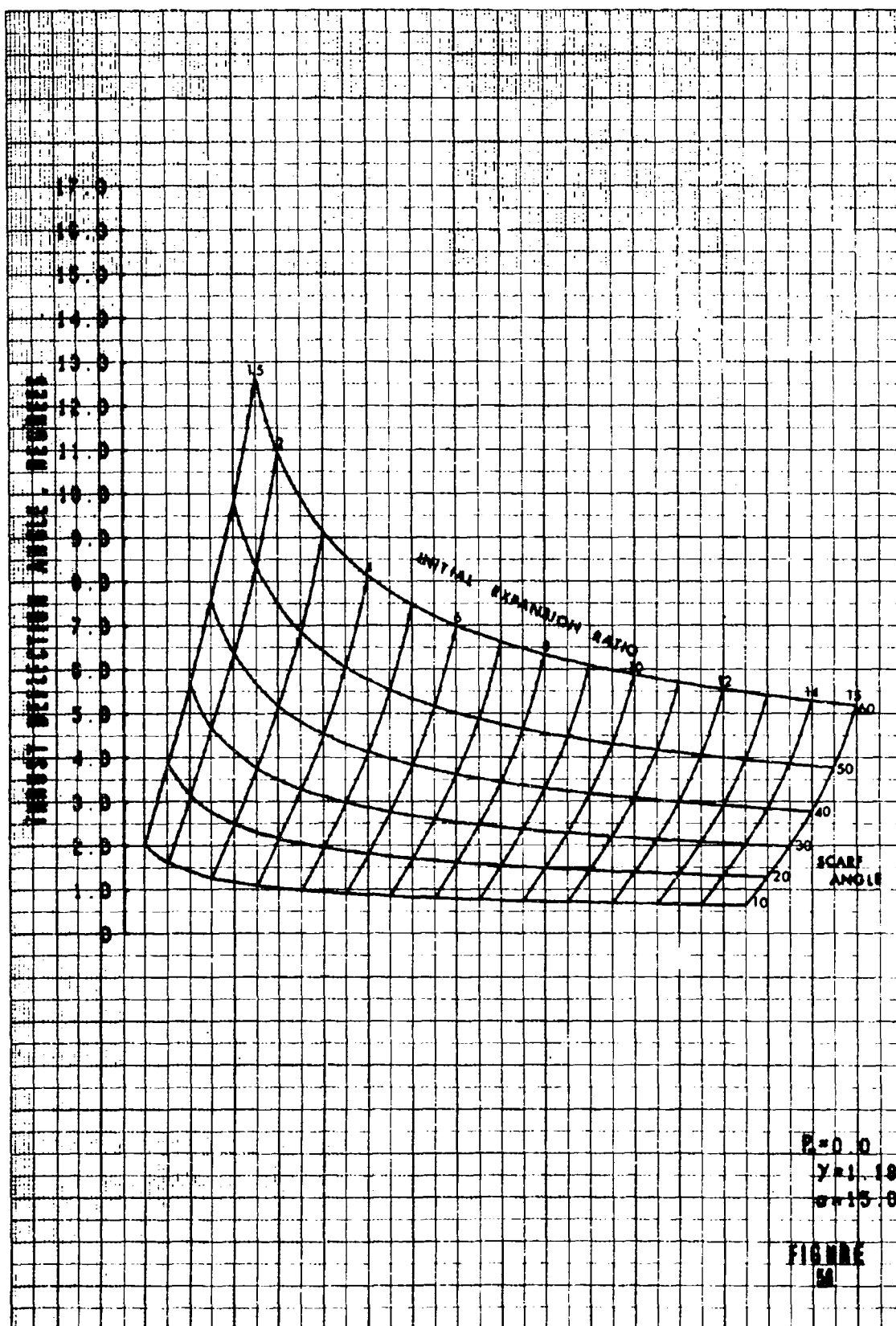


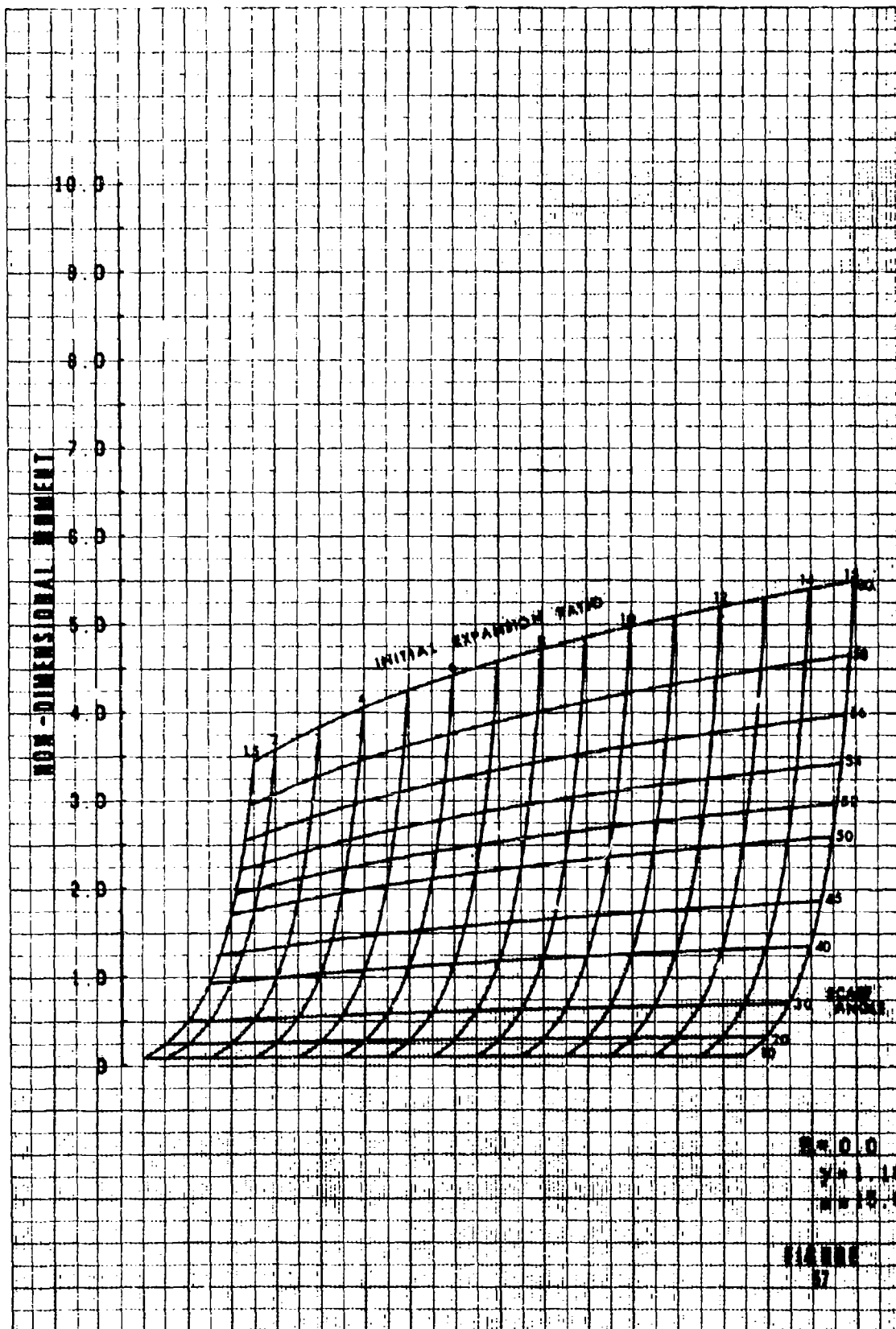












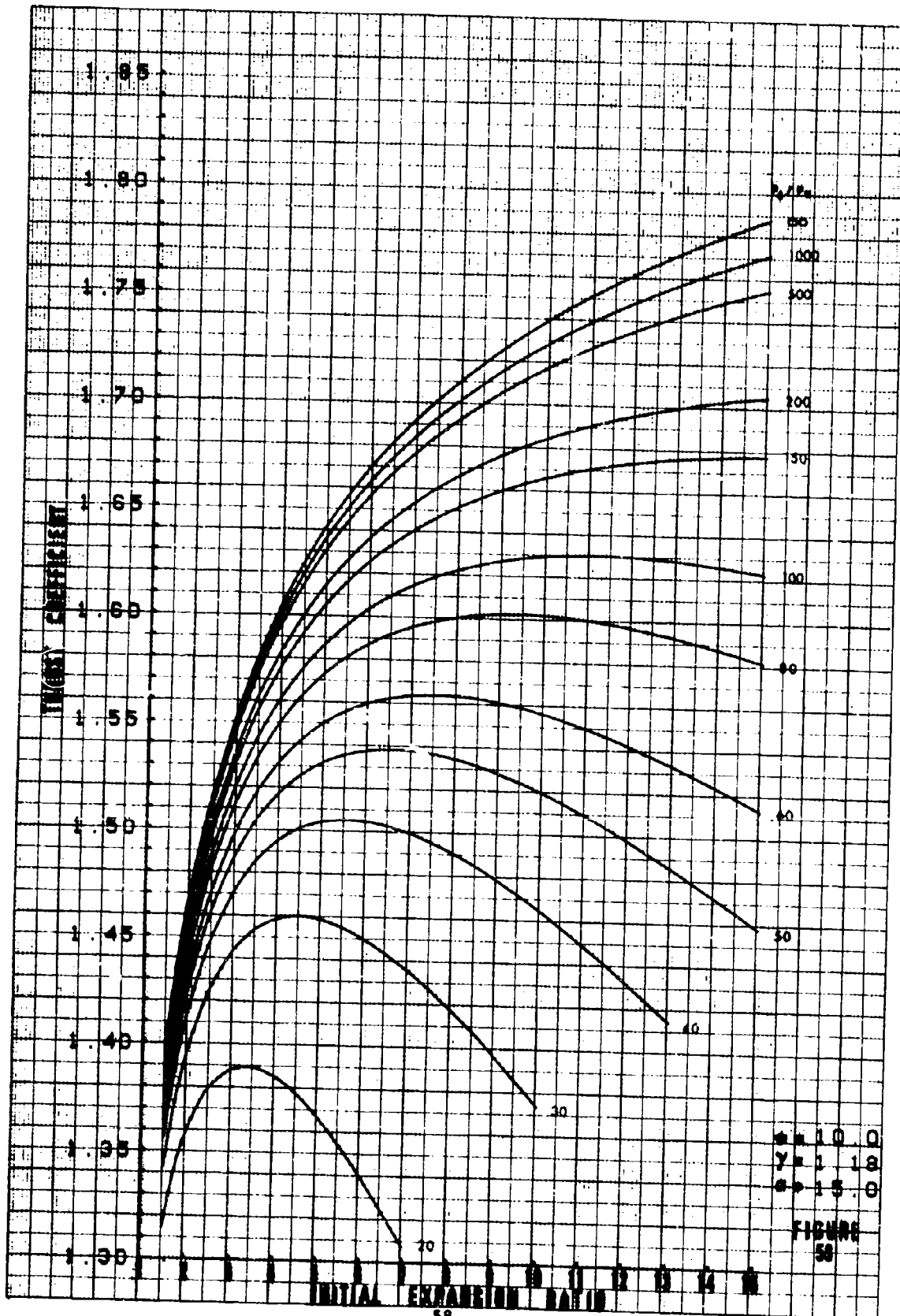
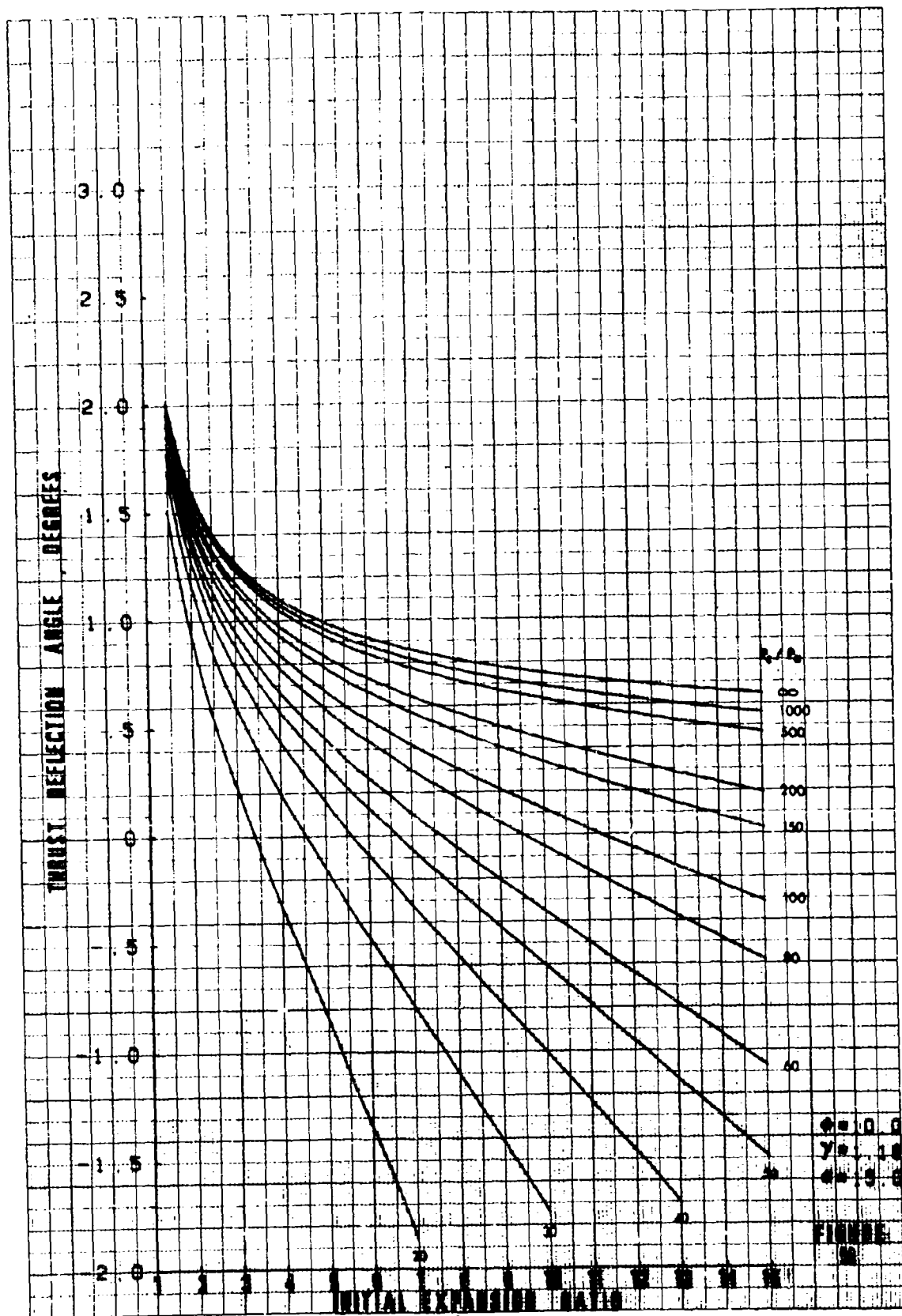
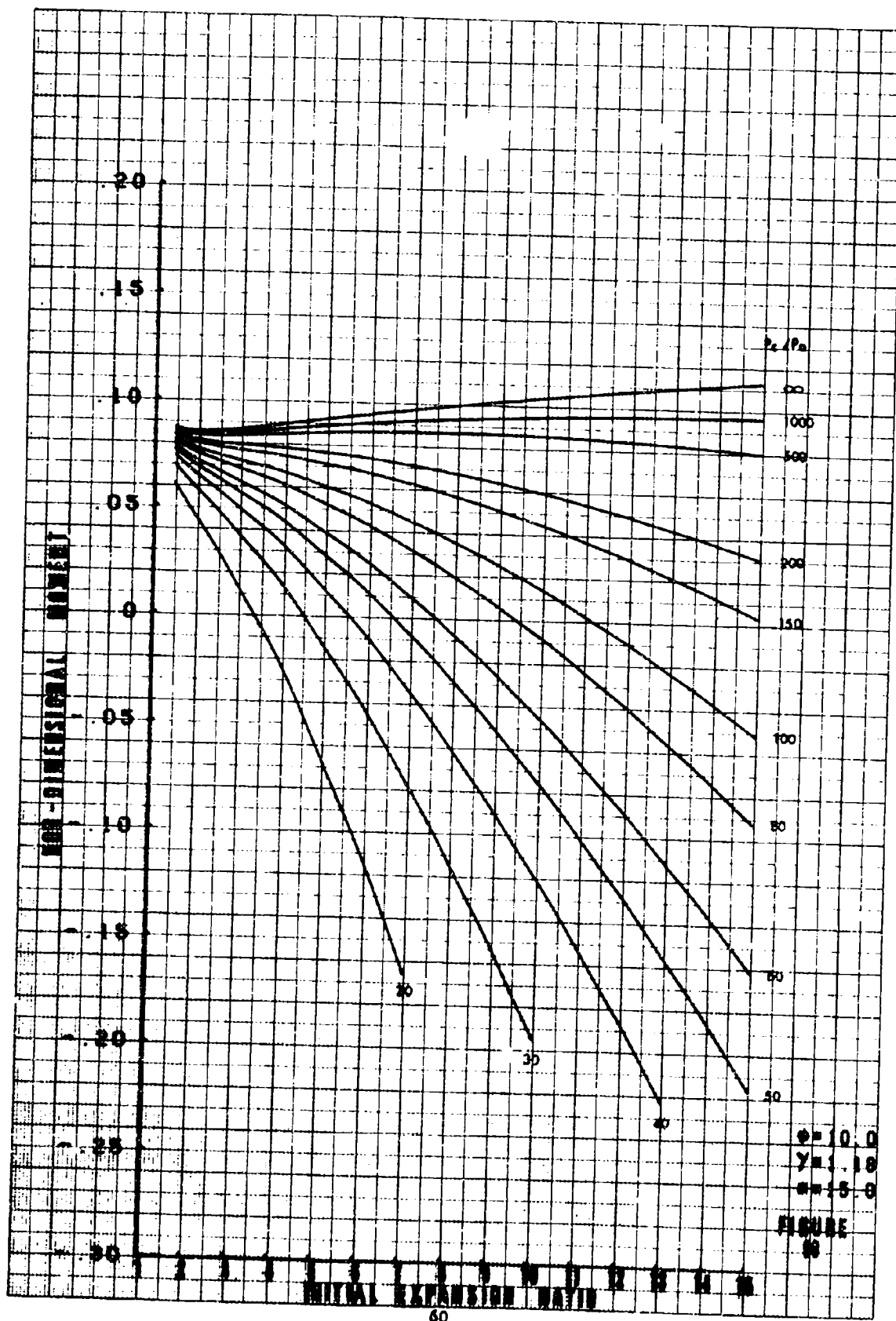


FIGURE 58





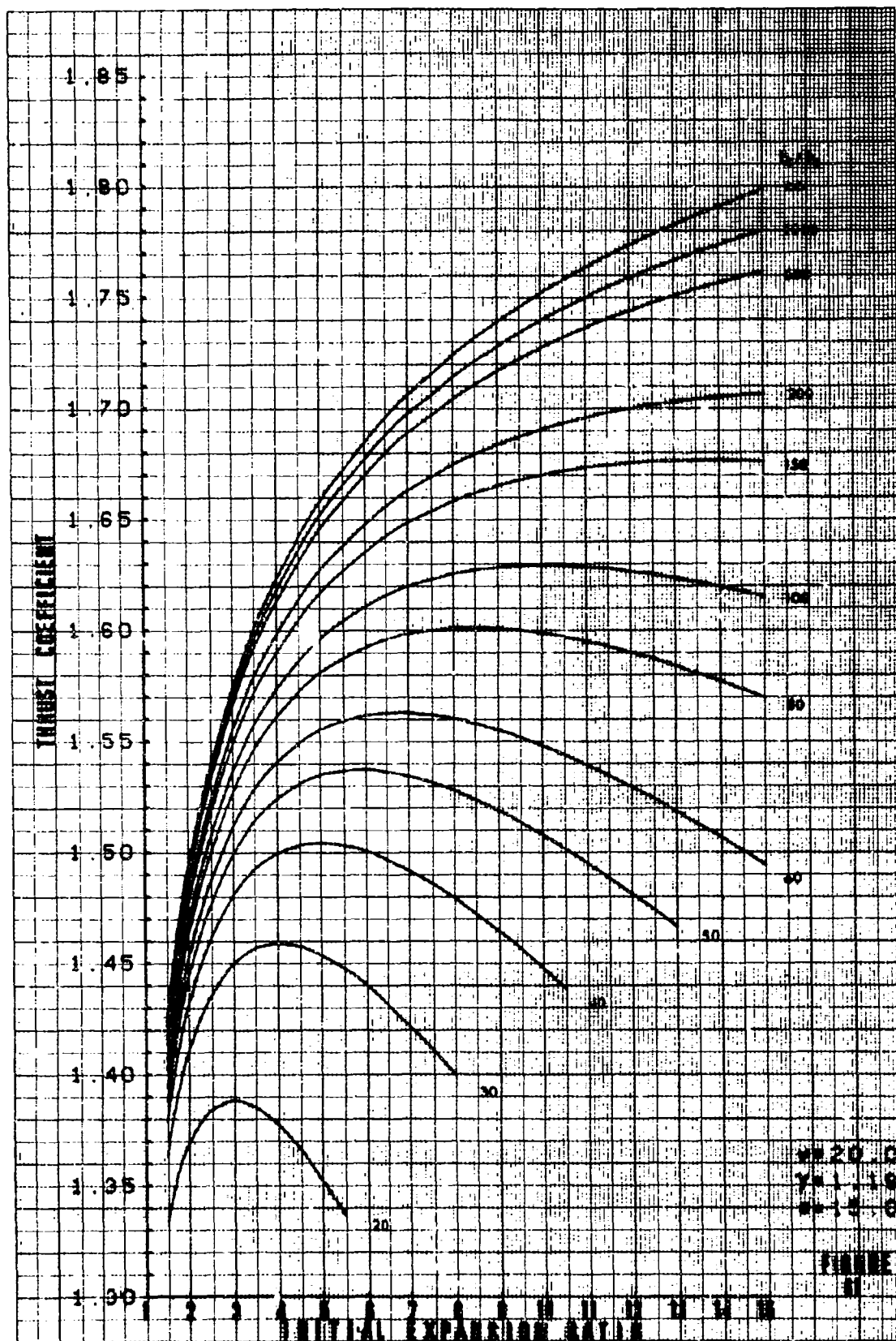
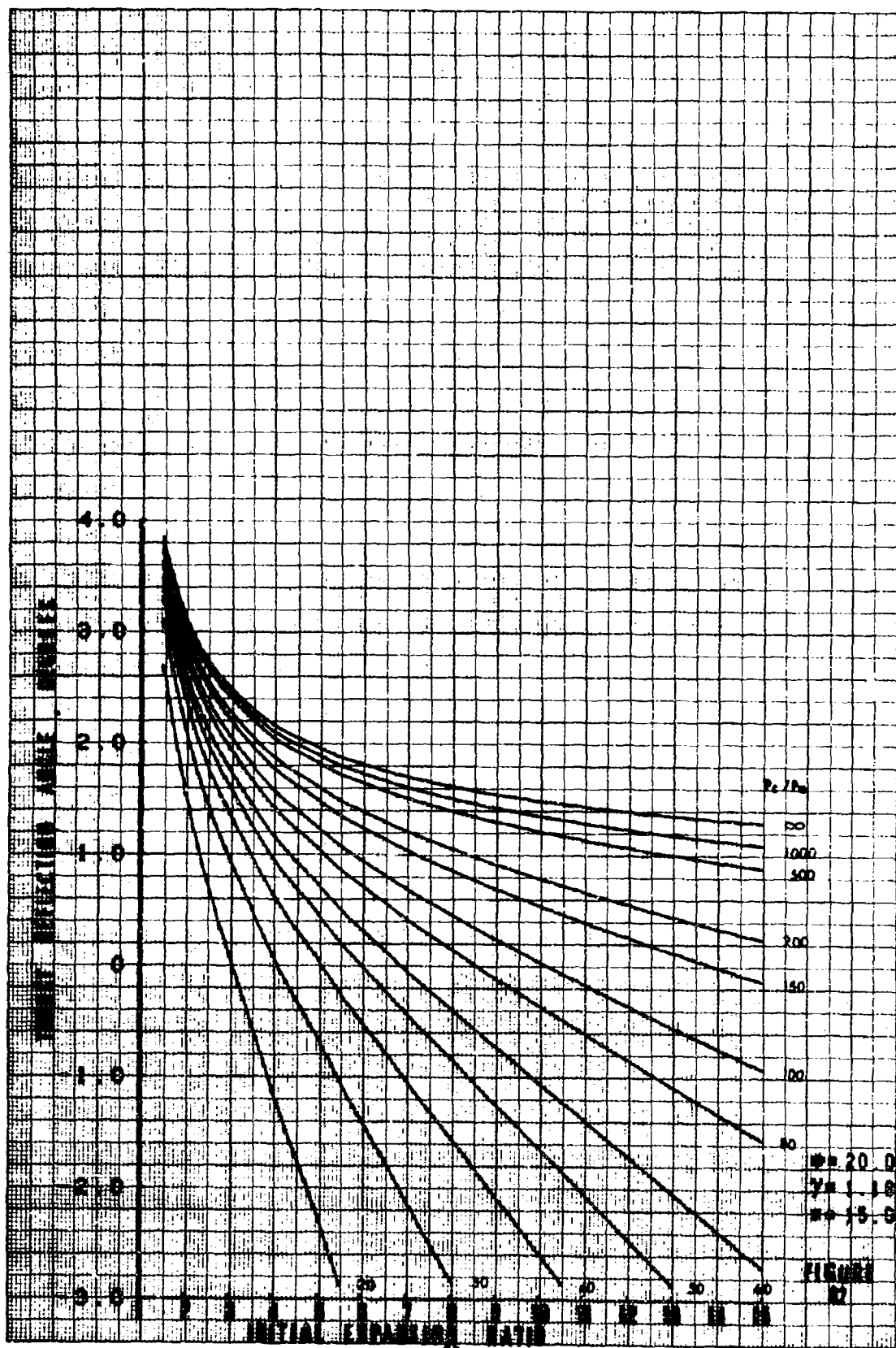
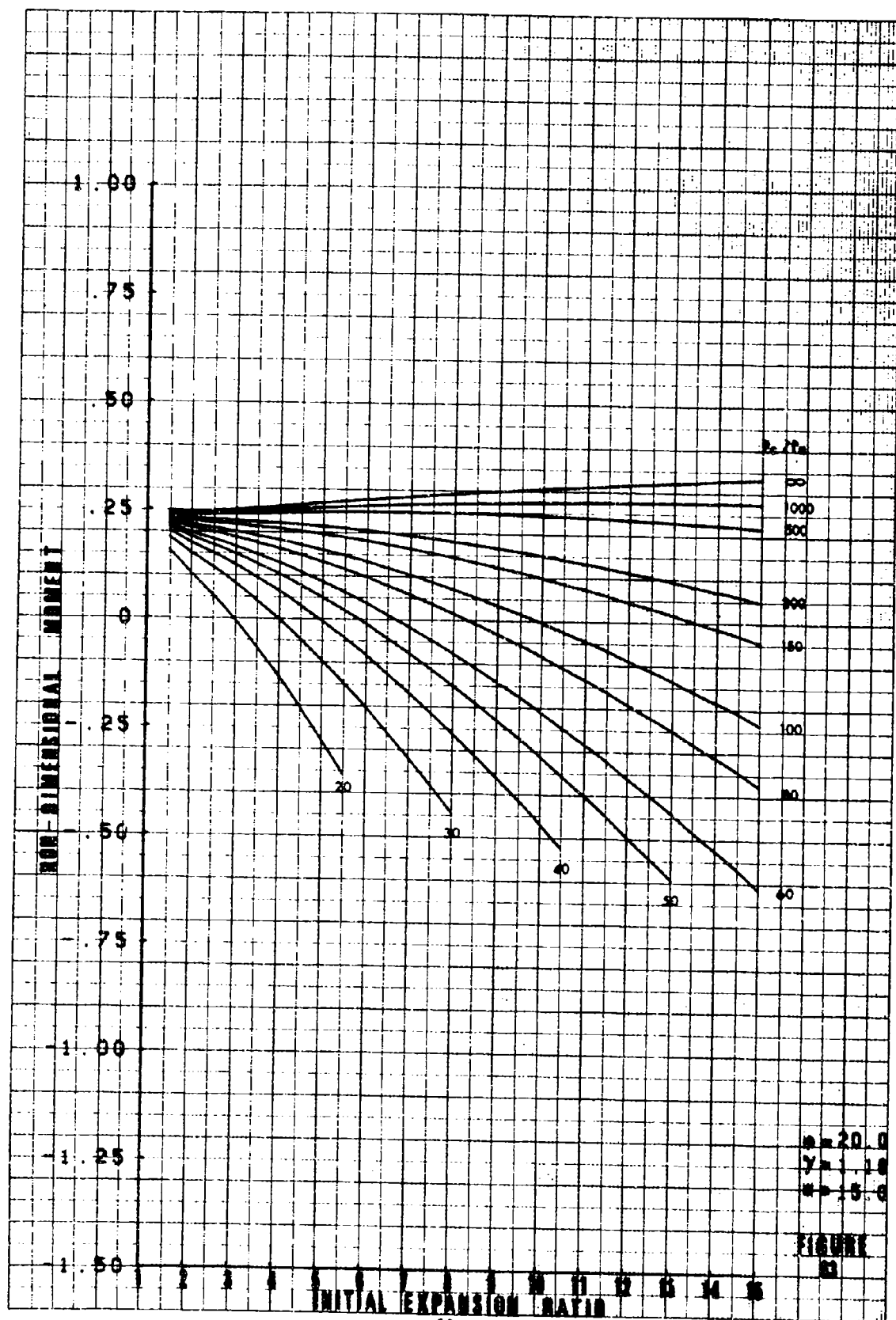


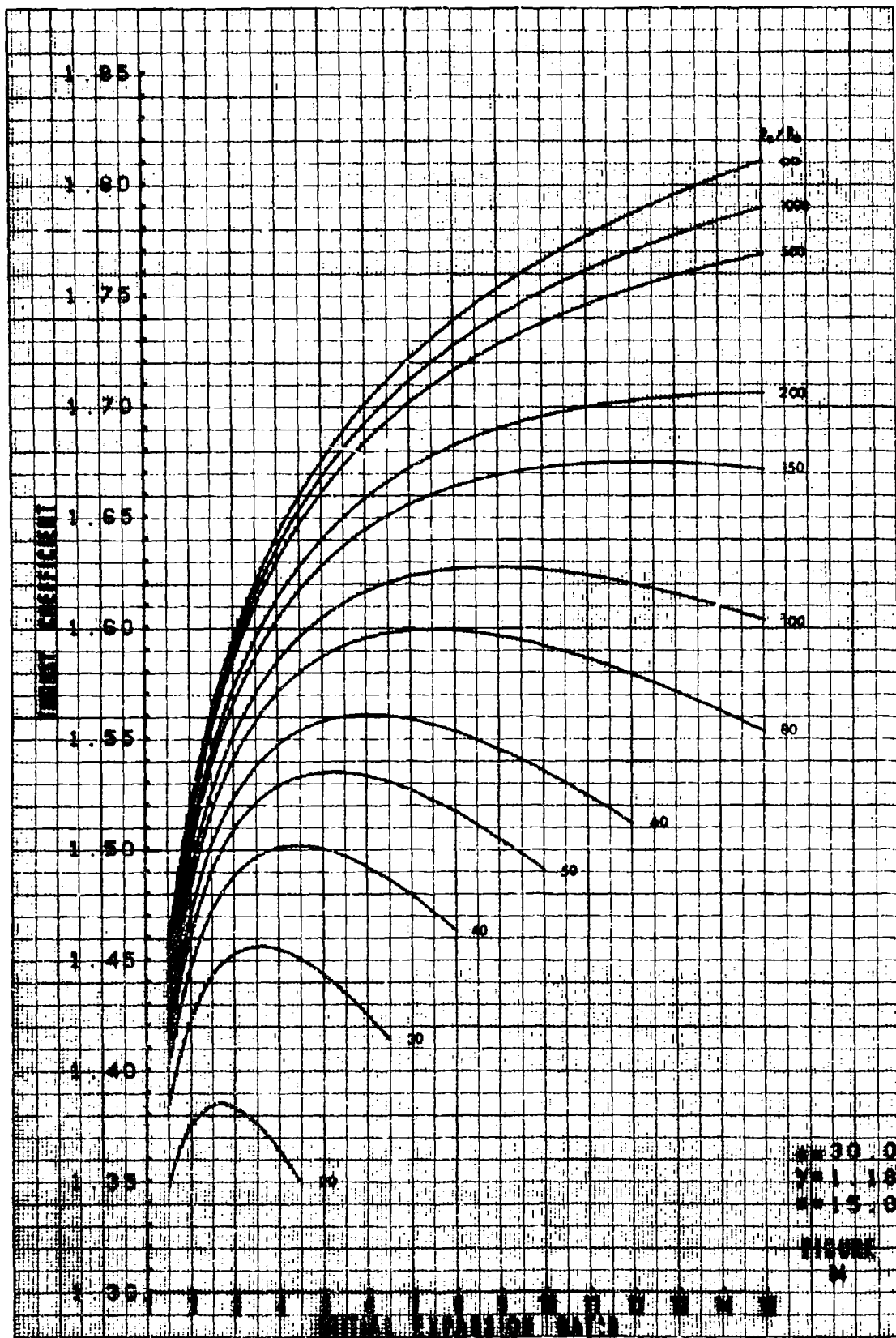
FIGURE 1

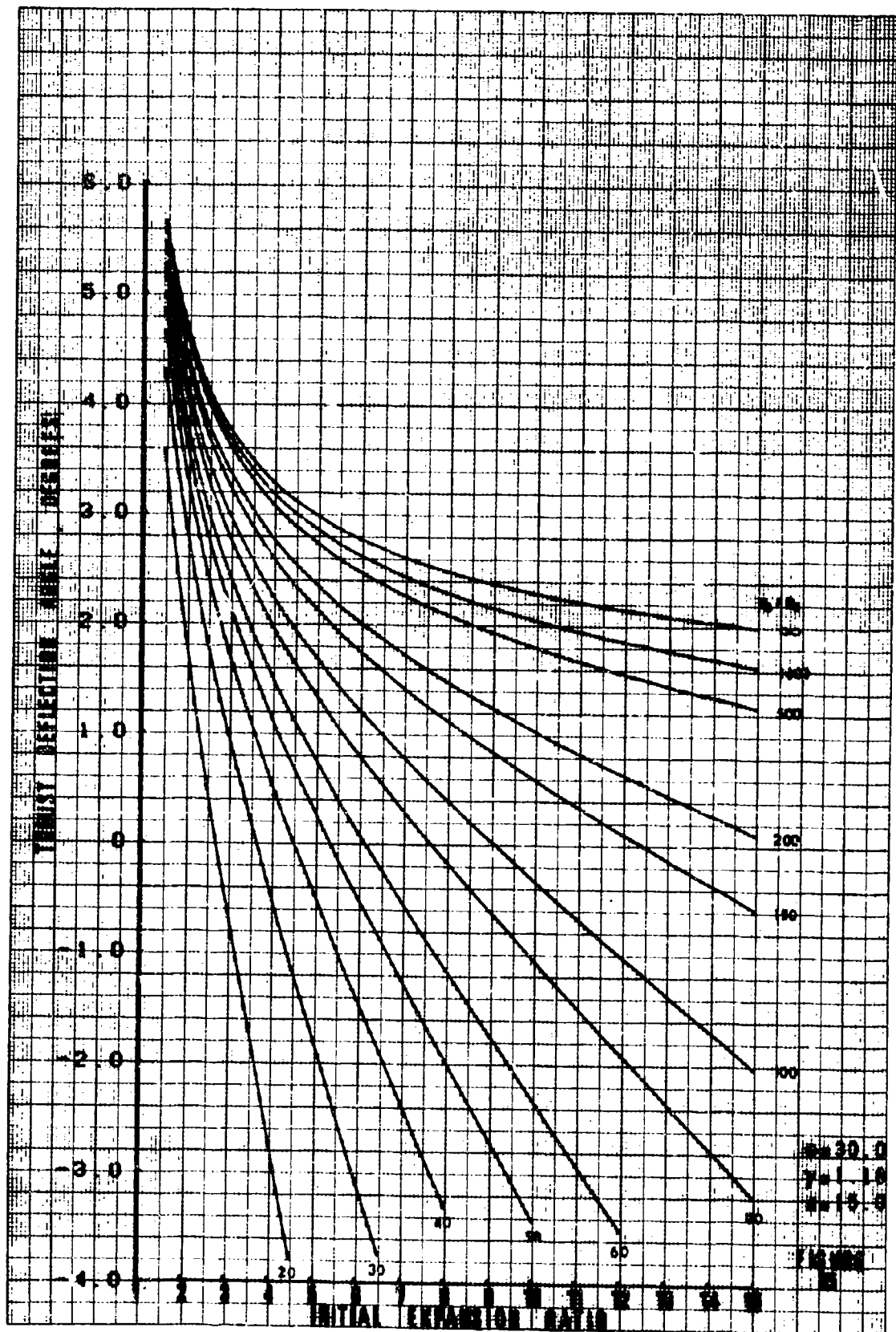


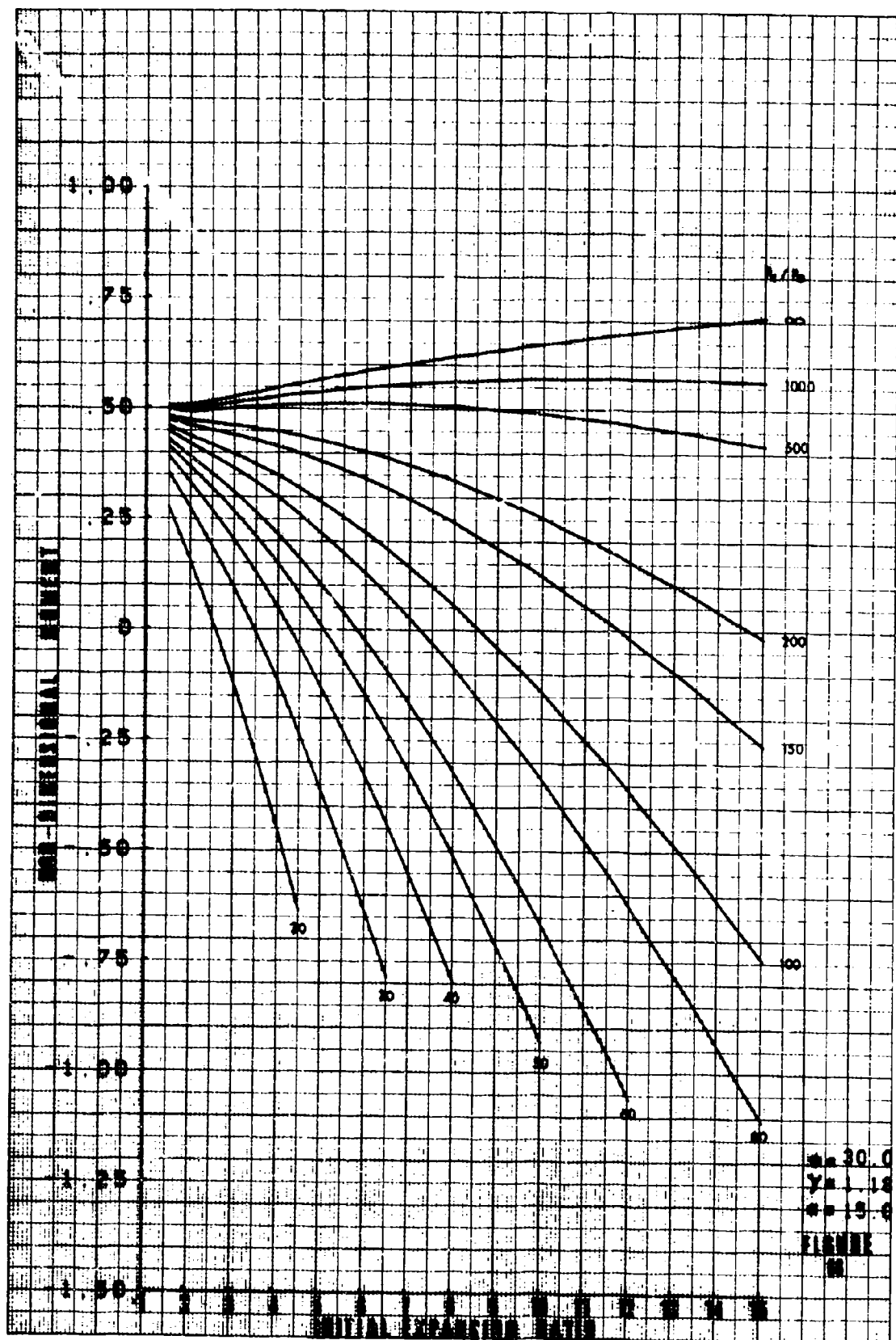


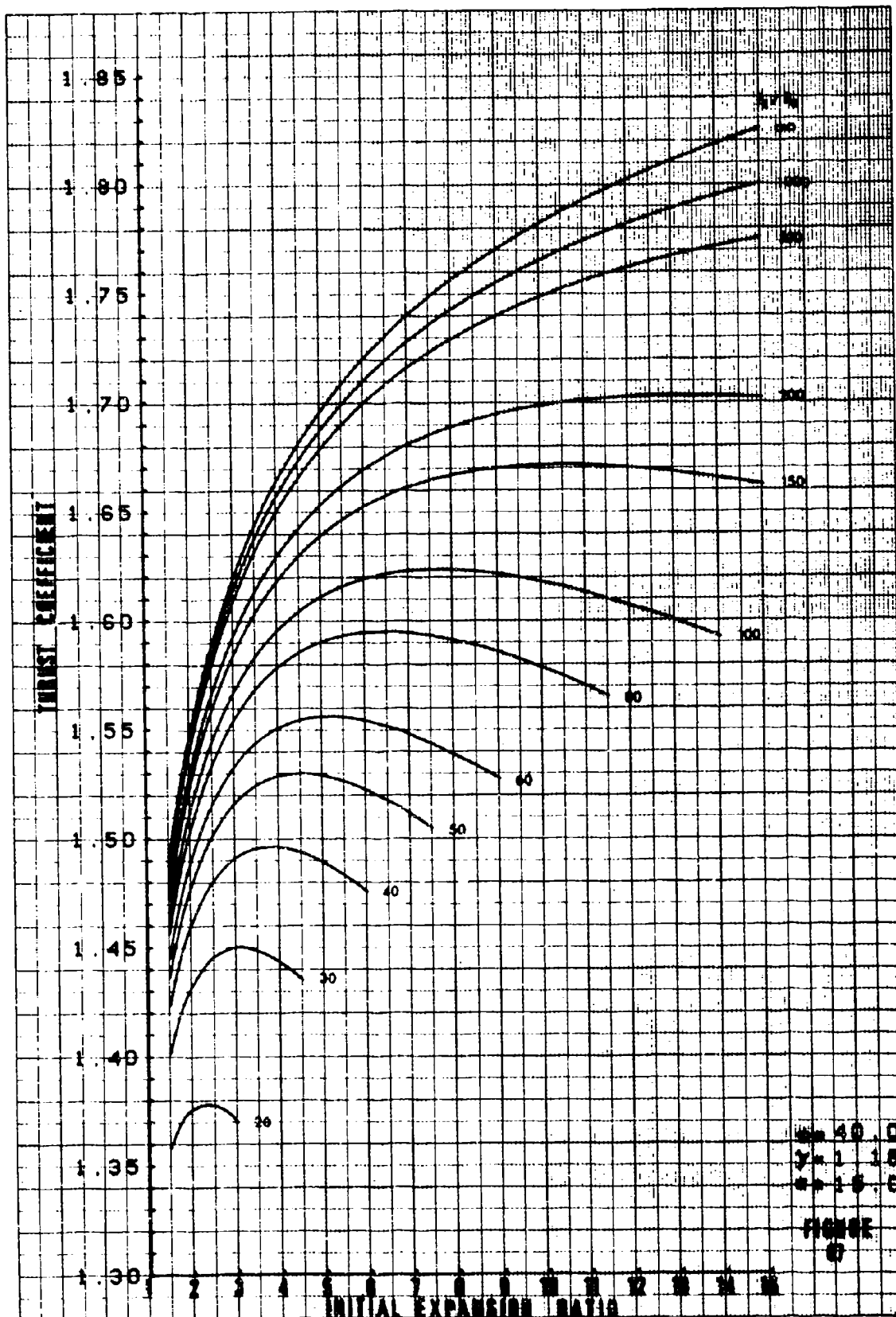
$\mu = 20.0$
 $\gamma = 1.10$
 $\kappa = 15.0$

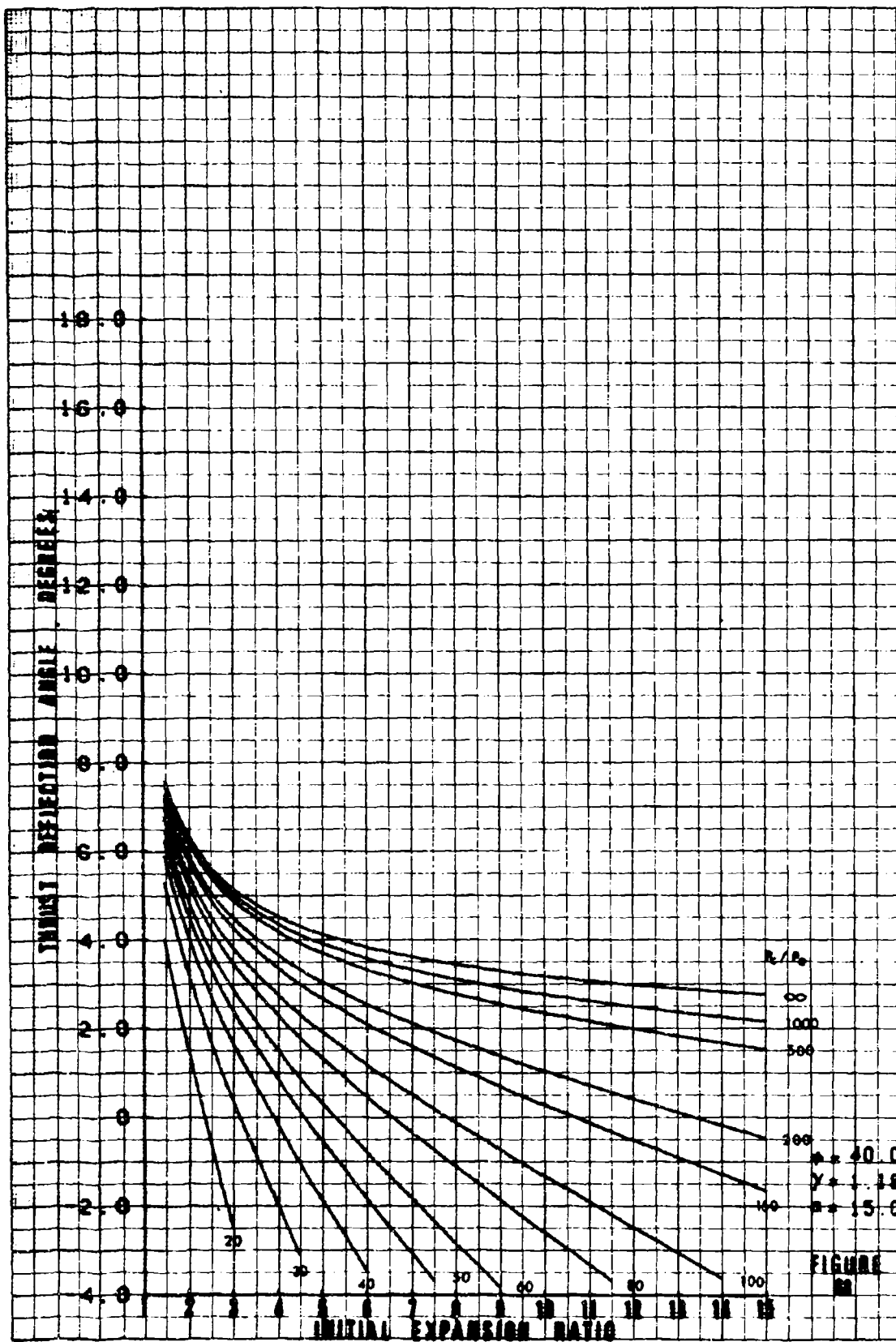
FIGURE 63

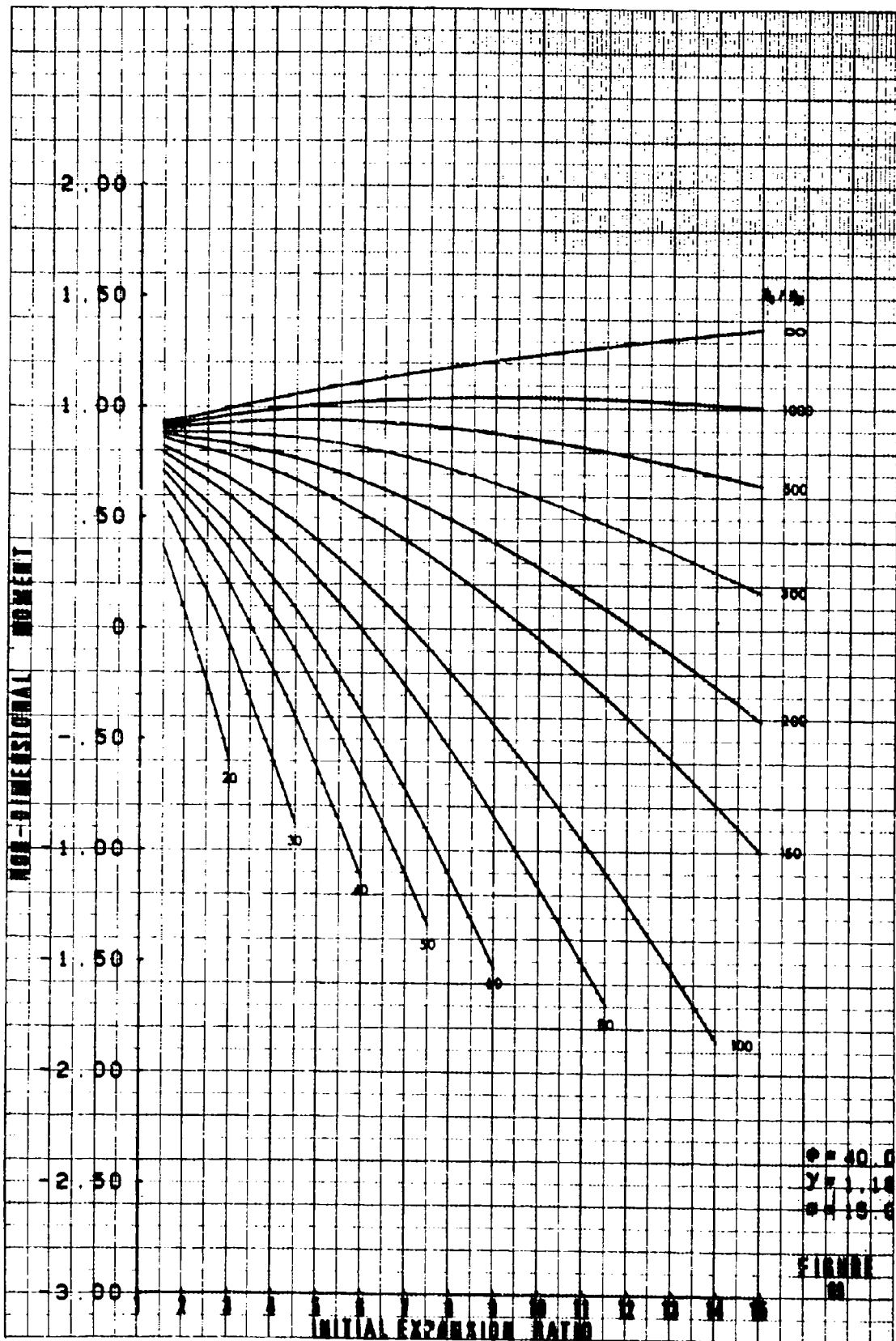






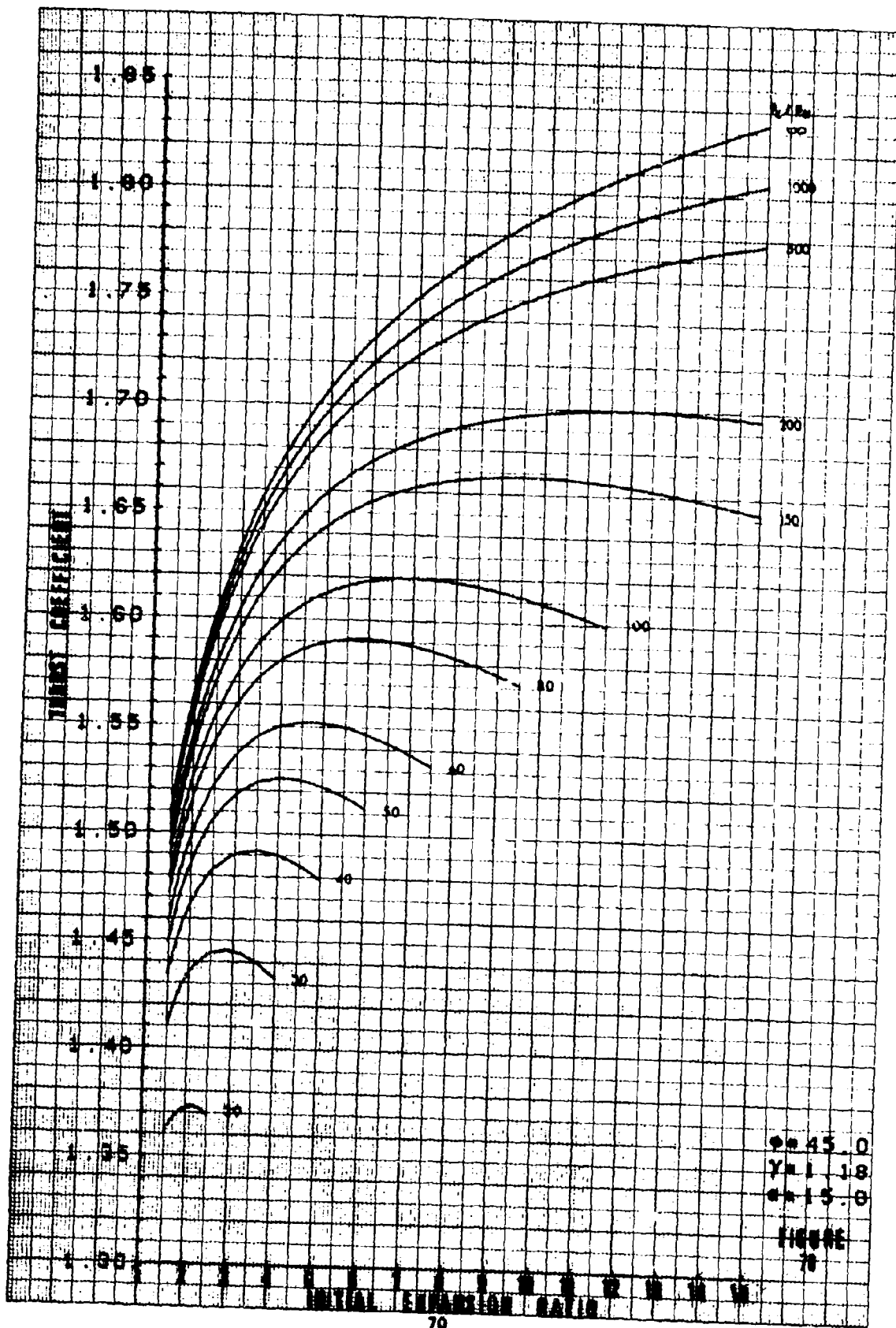






$\phi = 40.0$
 $\gamma = 1.18$
 $\sigma = 19.6$

FIGURE 88



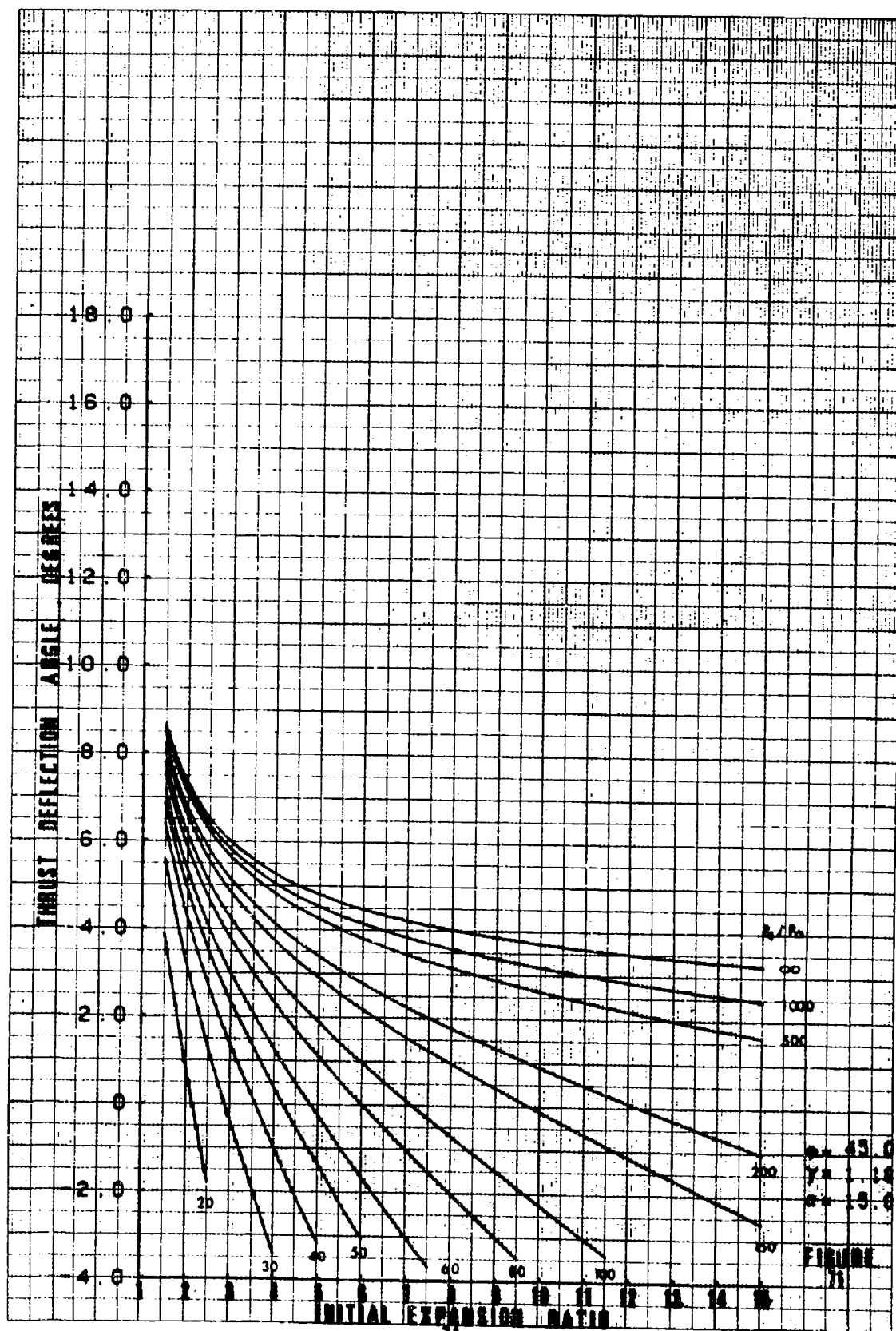
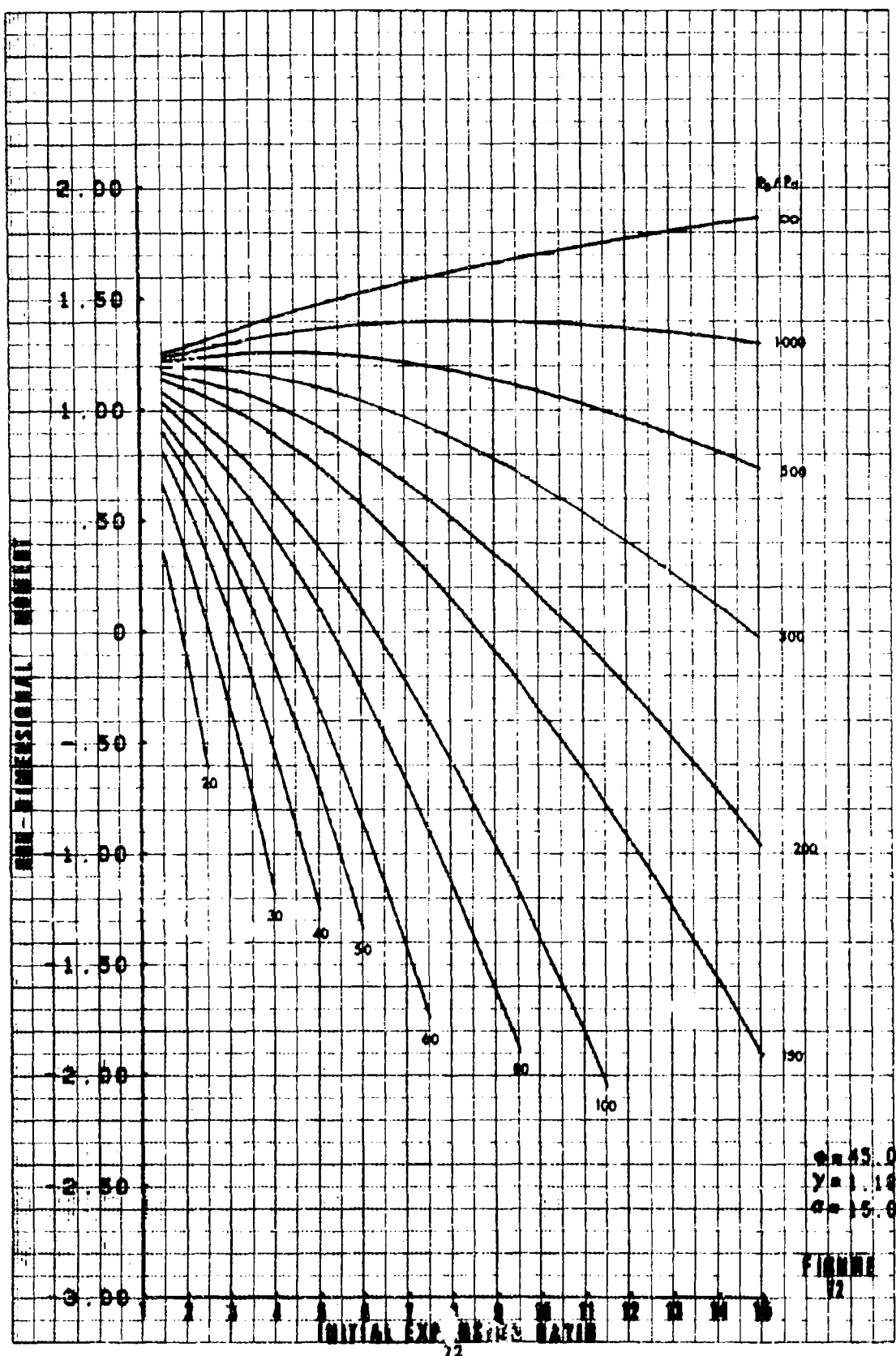
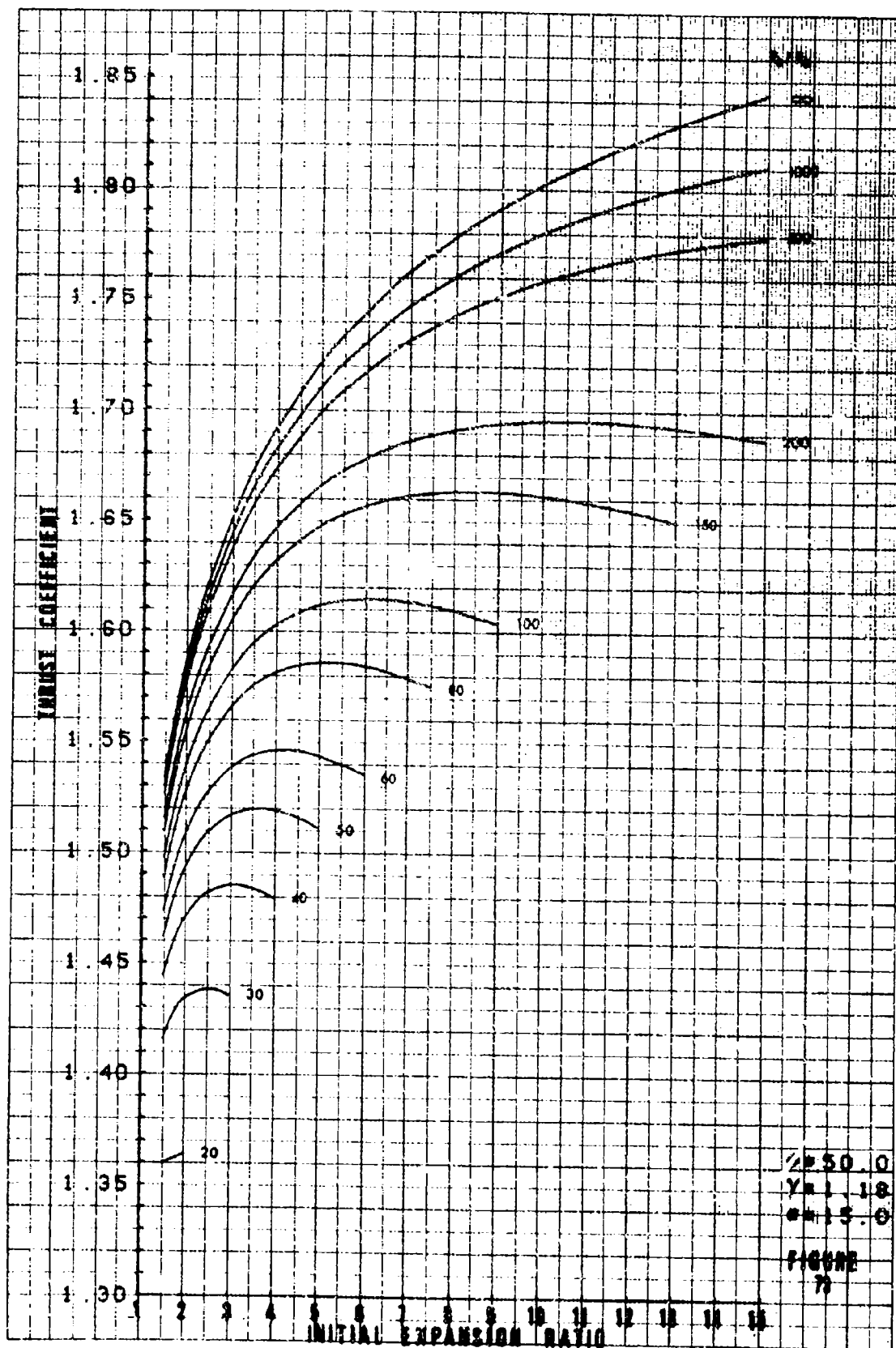


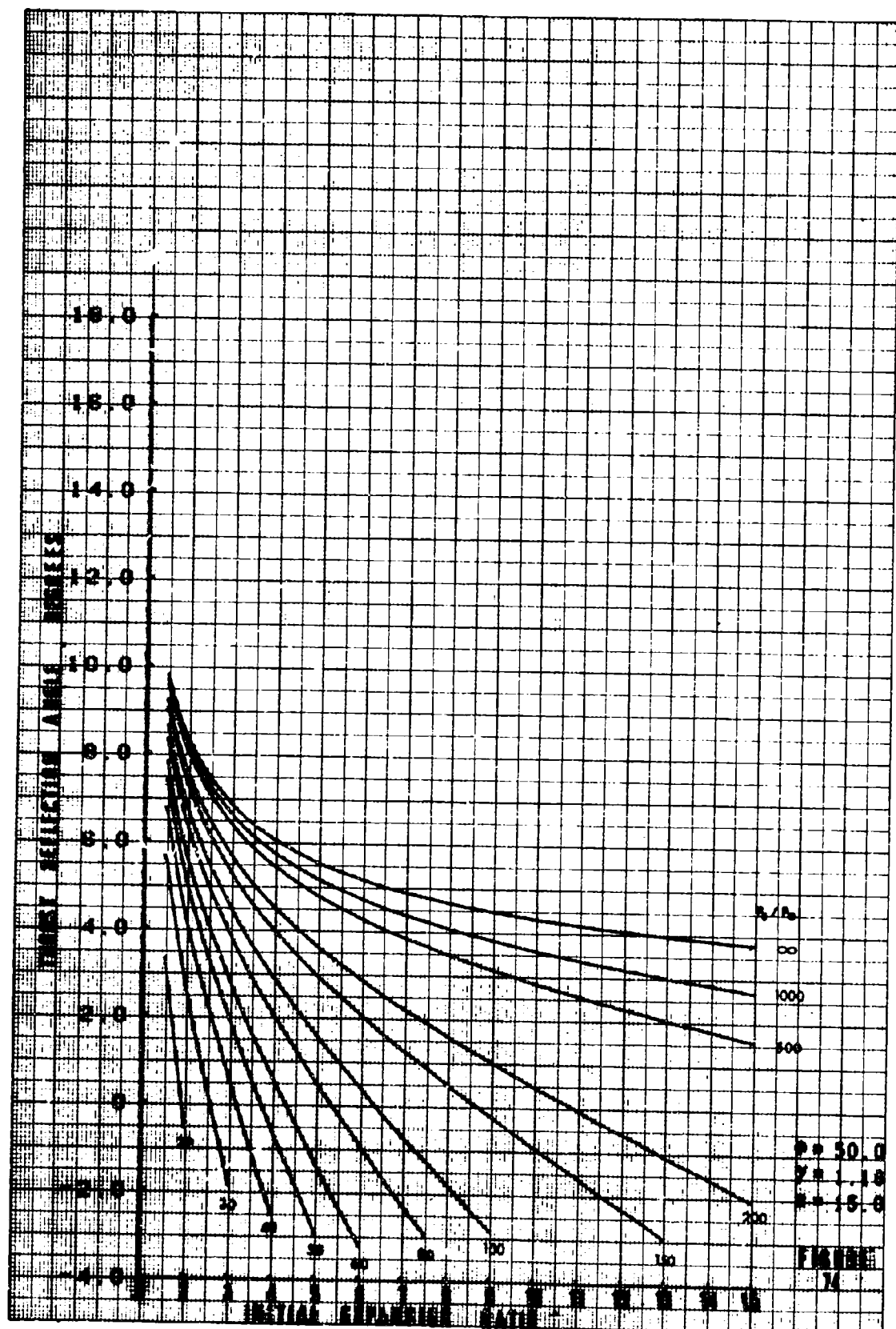
FIGURE 71

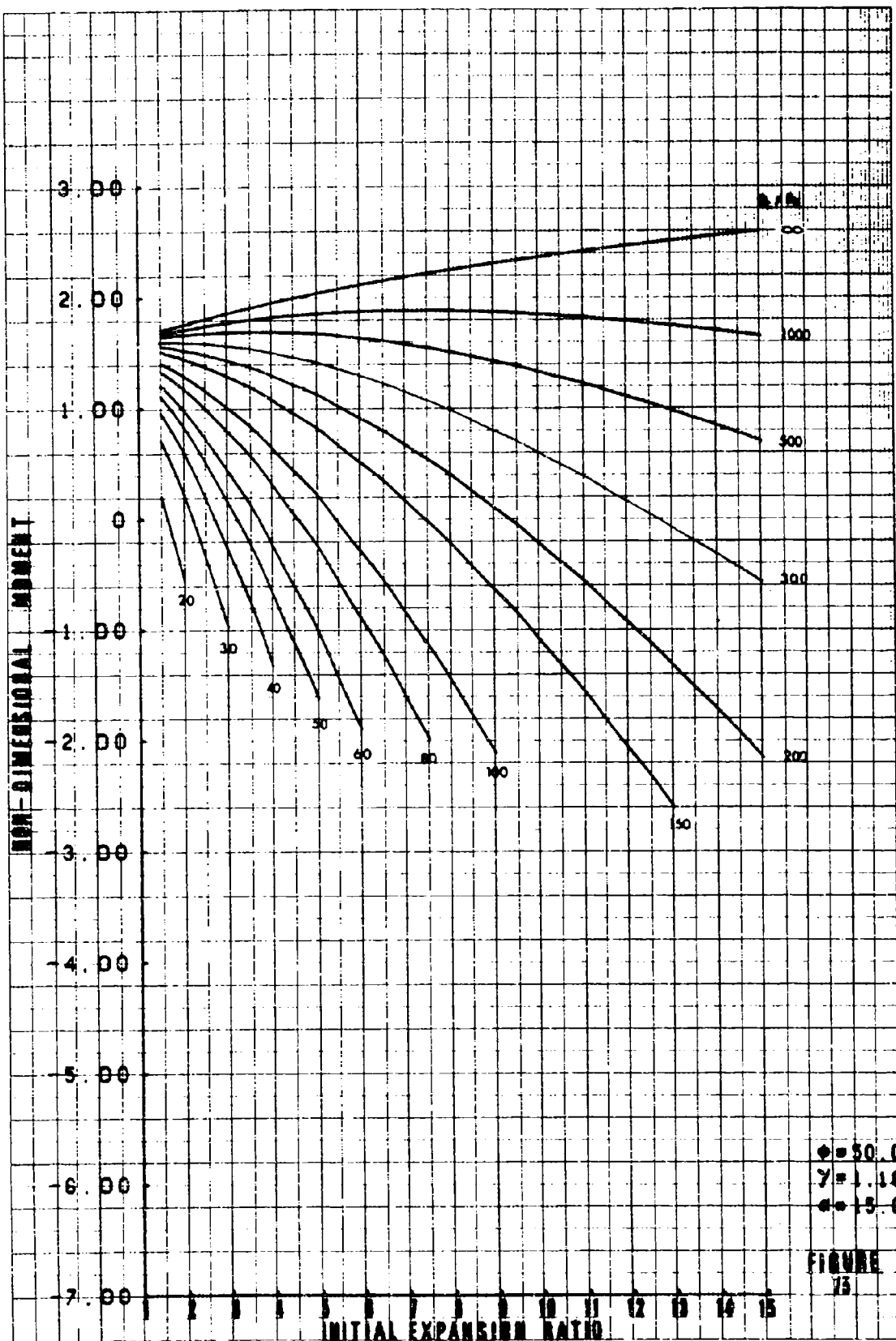


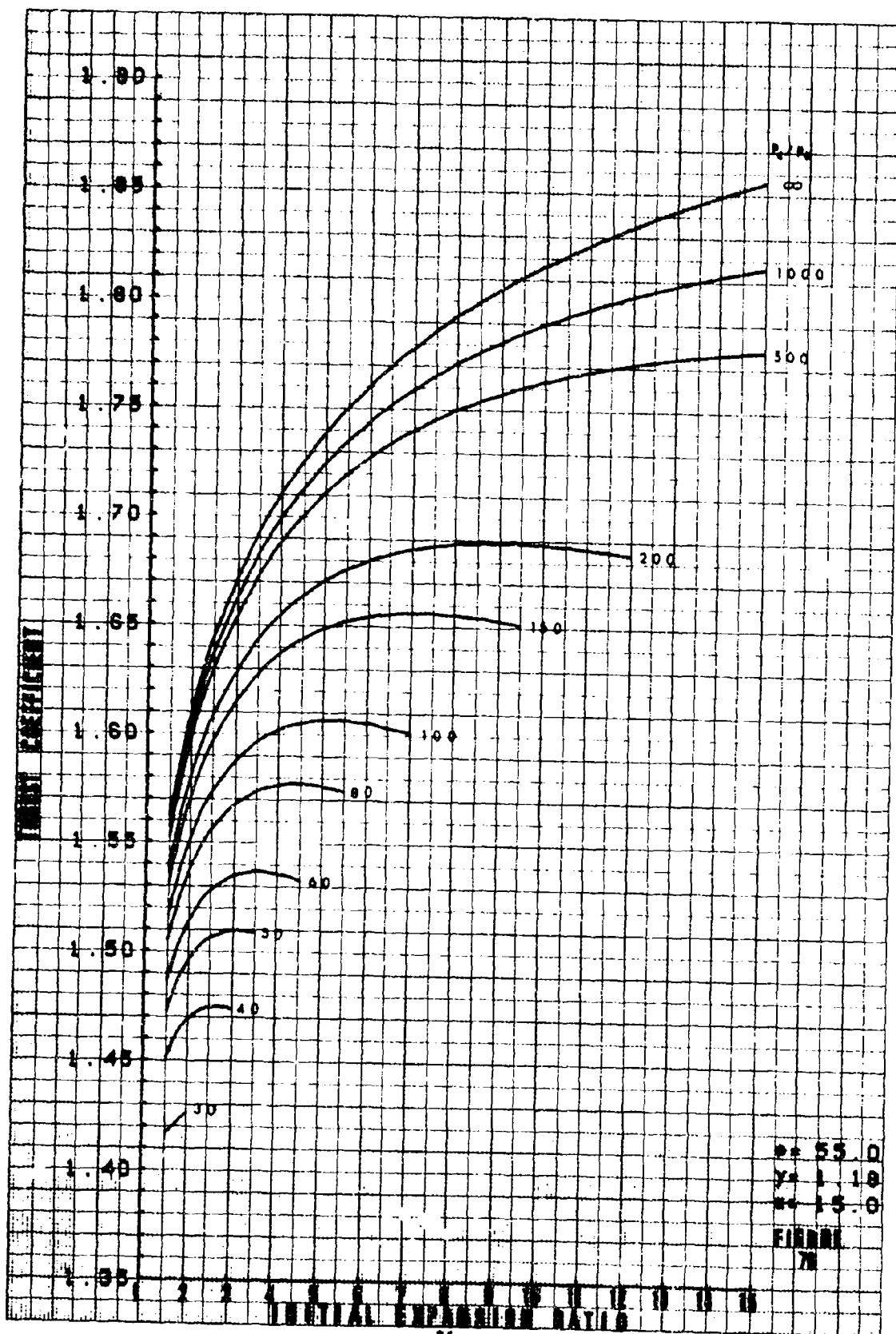


$\gamma = 50.0$
 $Y = 1.18$
 $\phi = 15.0$

FIGURE 7







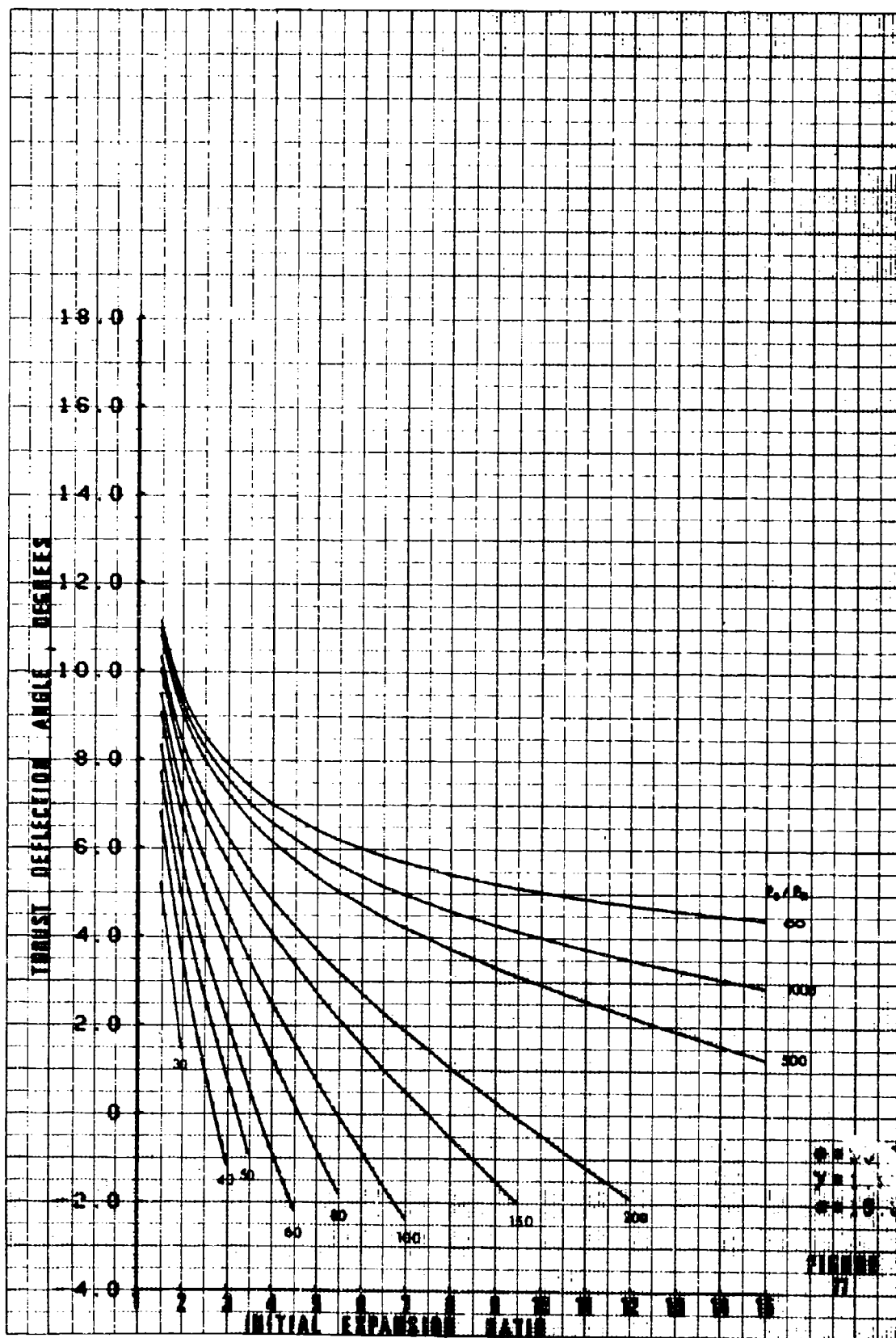
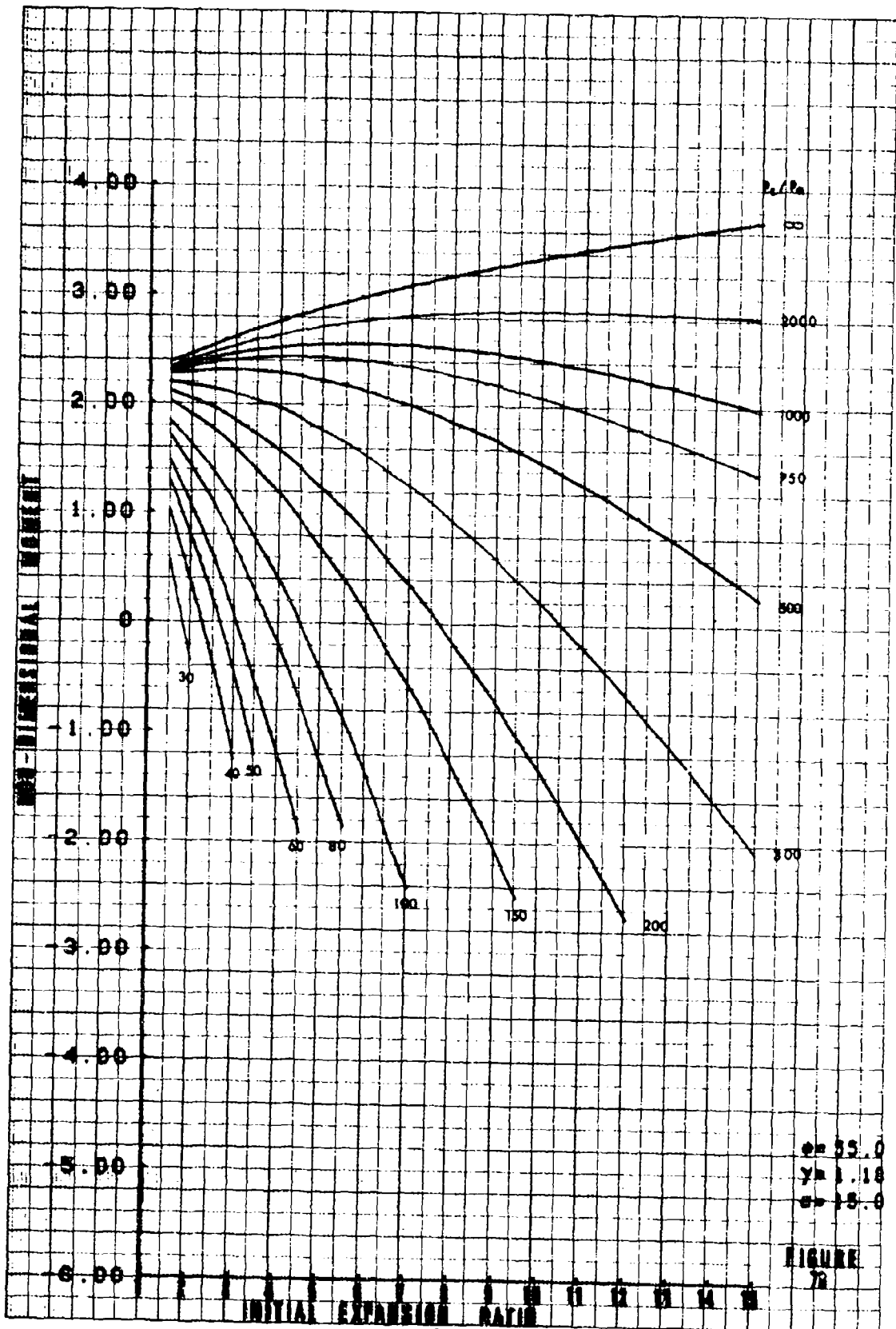
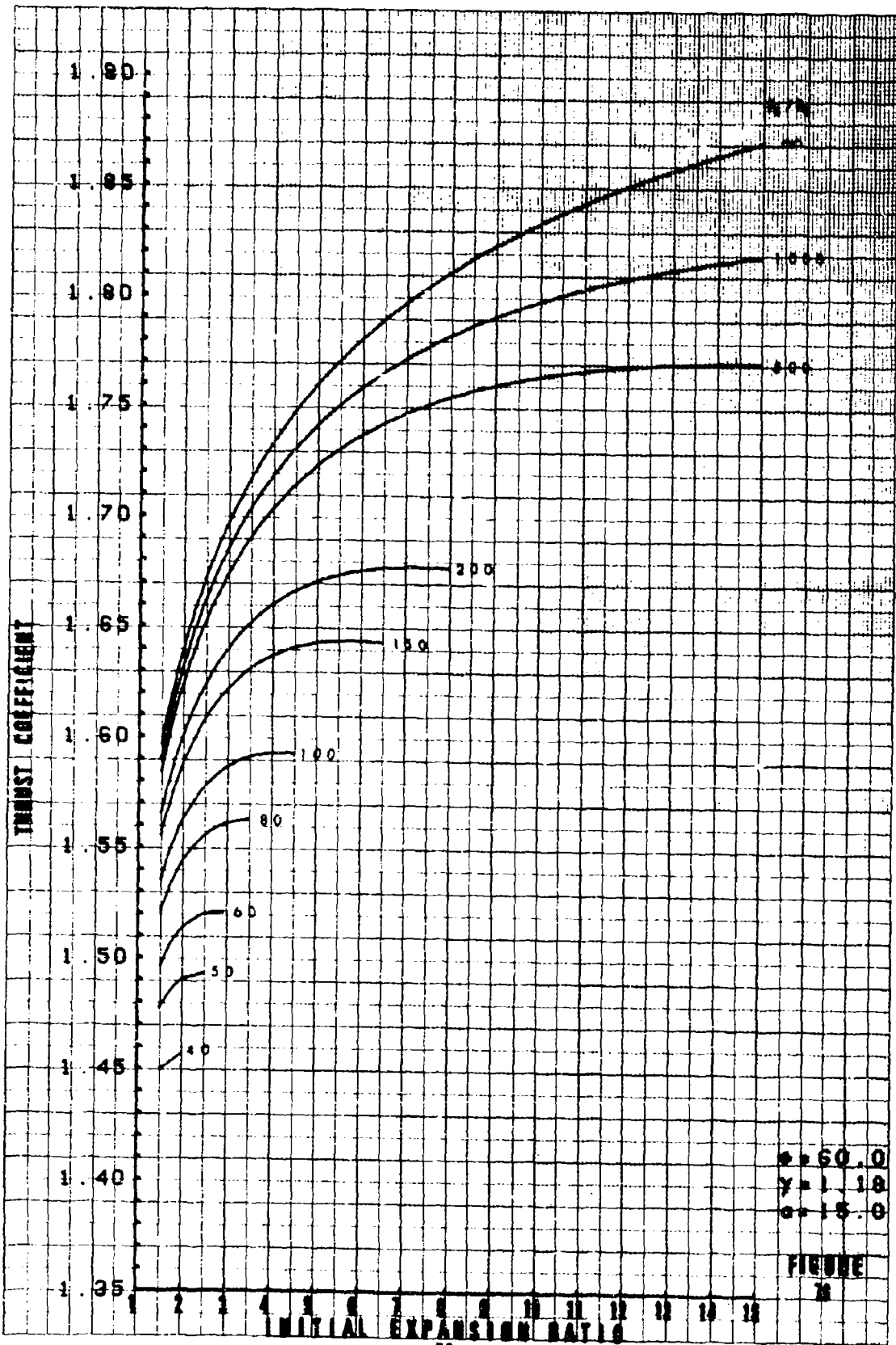


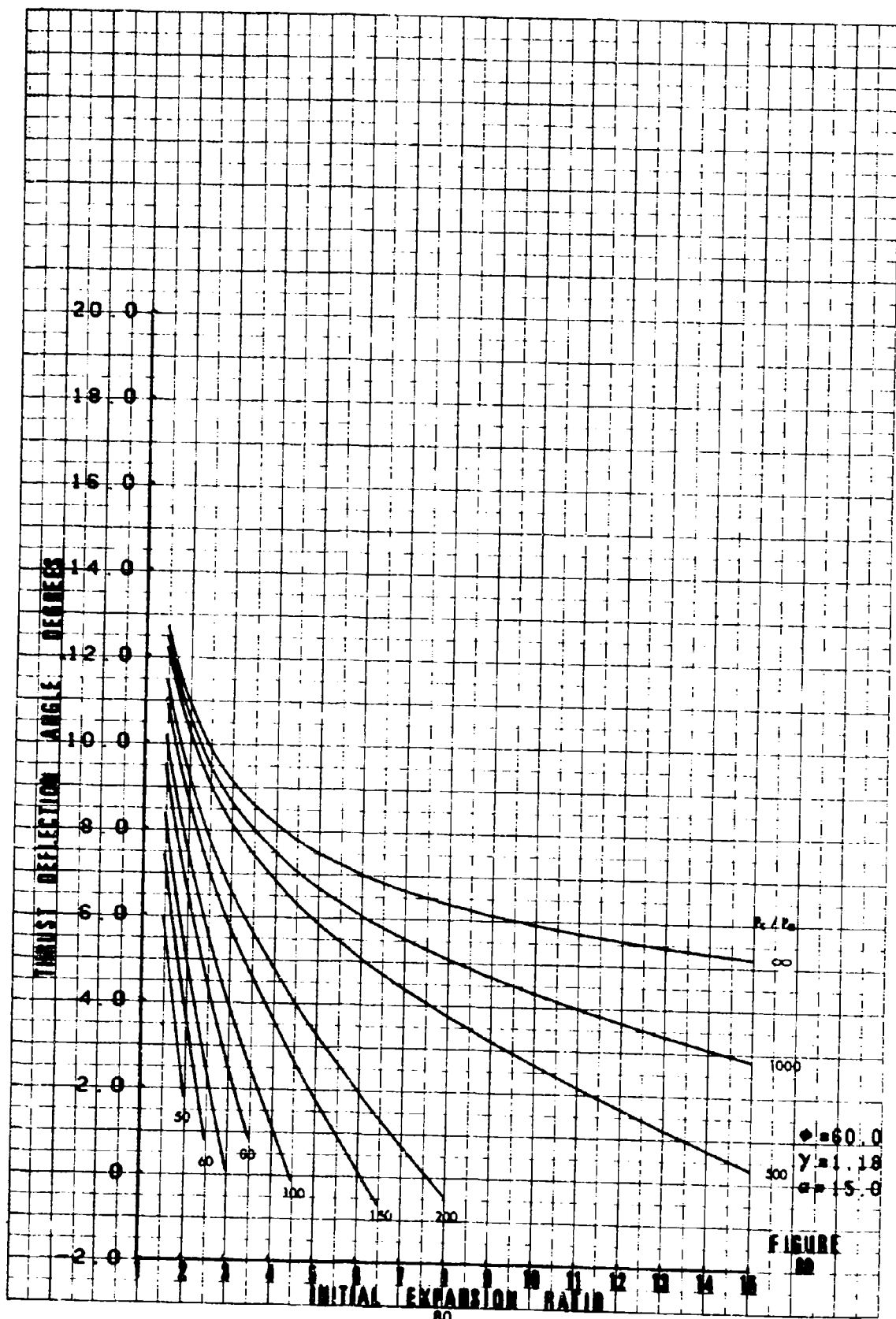
FIGURE 77

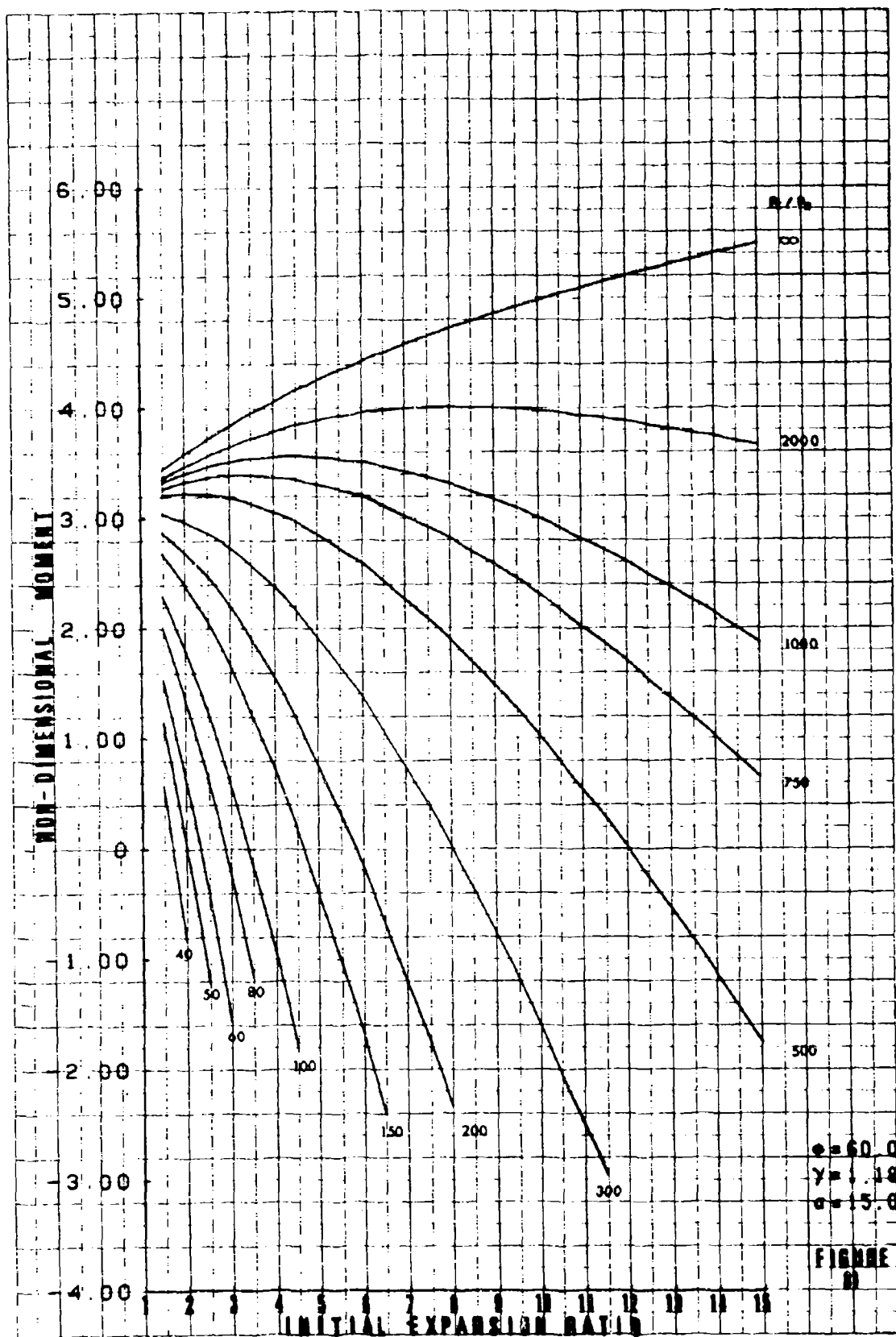


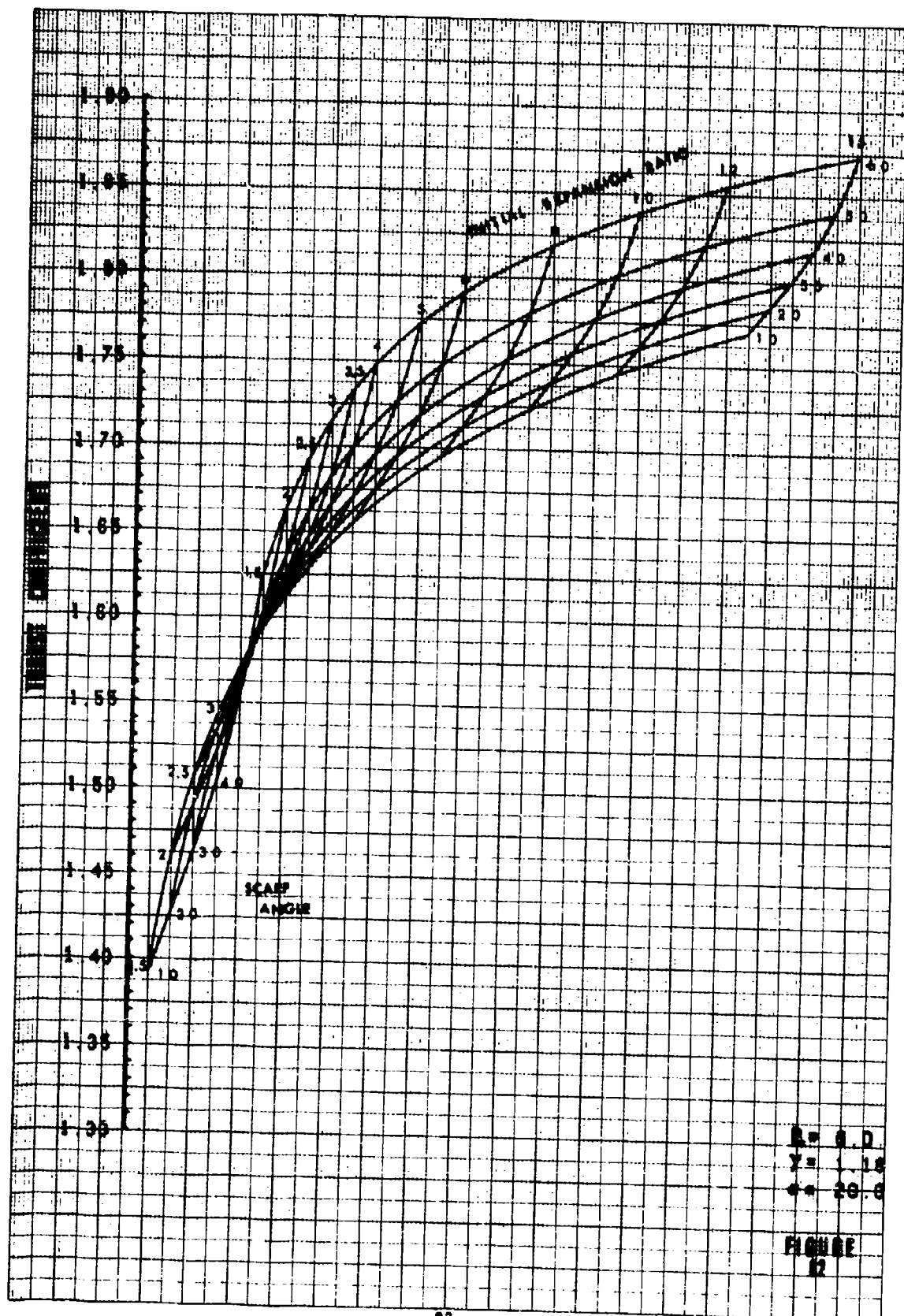


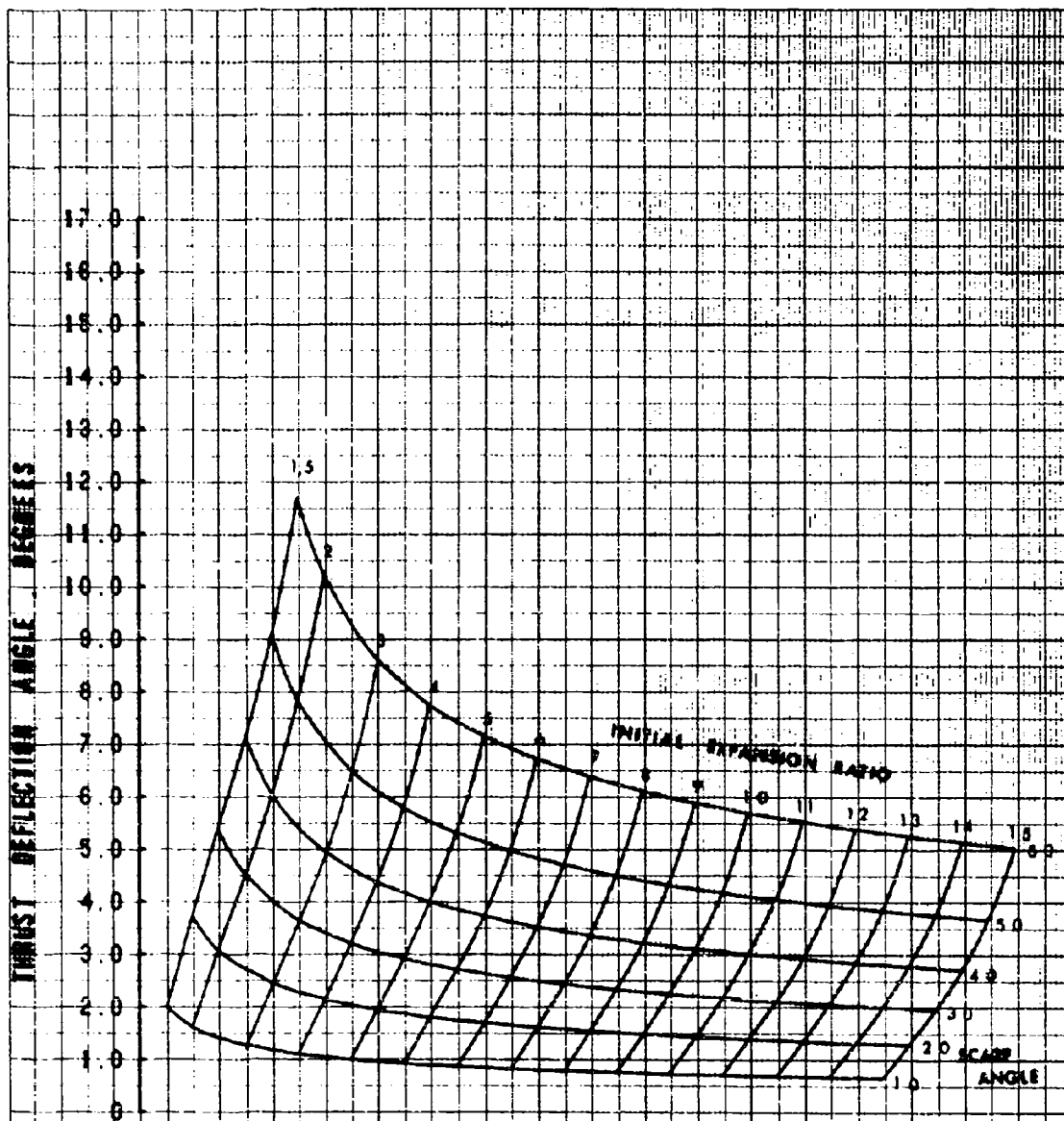
$\gamma = 1.18$
 $\alpha = 15.0$

FIGURE 26



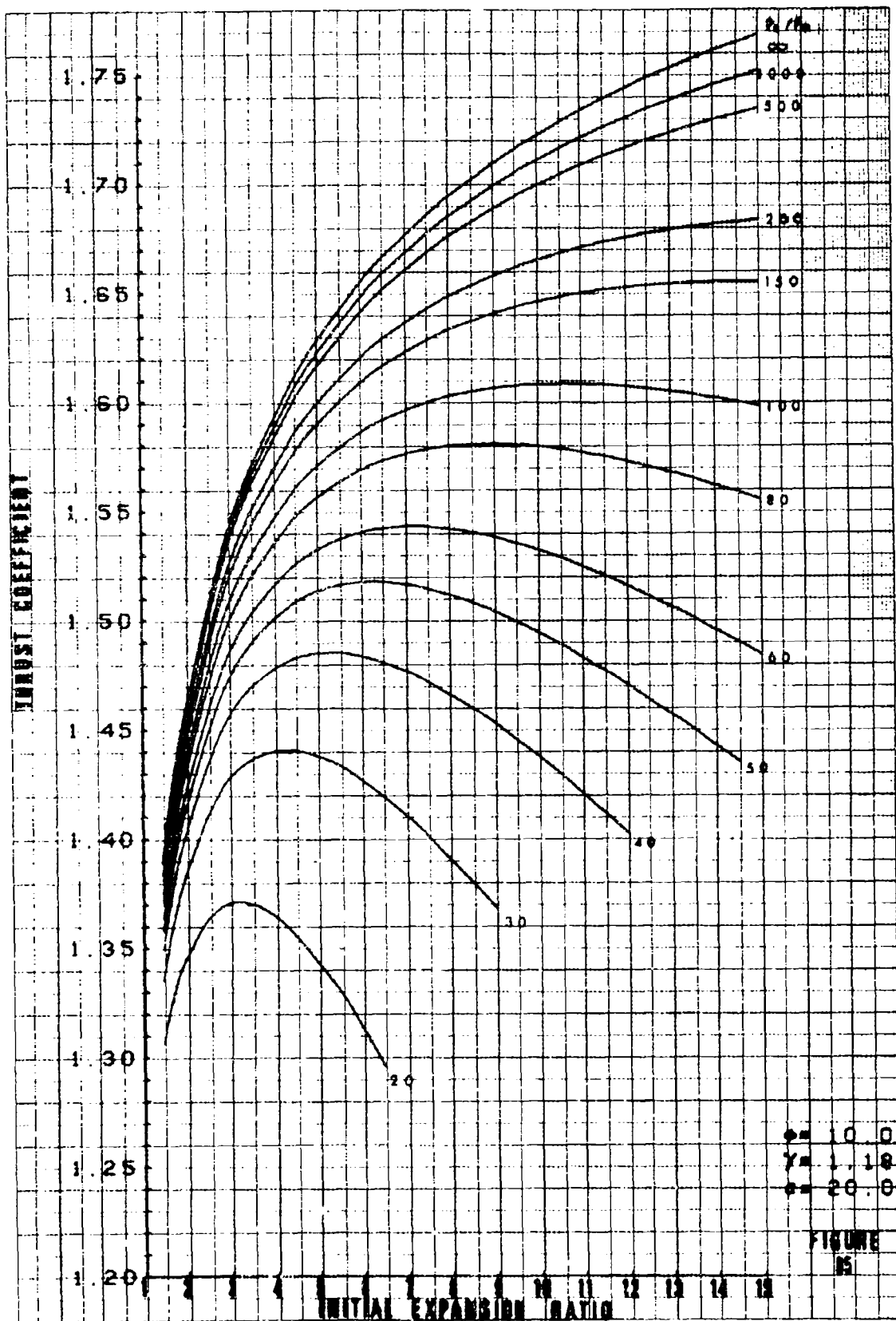






$R = 0.0$
 $\gamma = 1.18$
 $\sigma = 29.0$

FIGURE 10



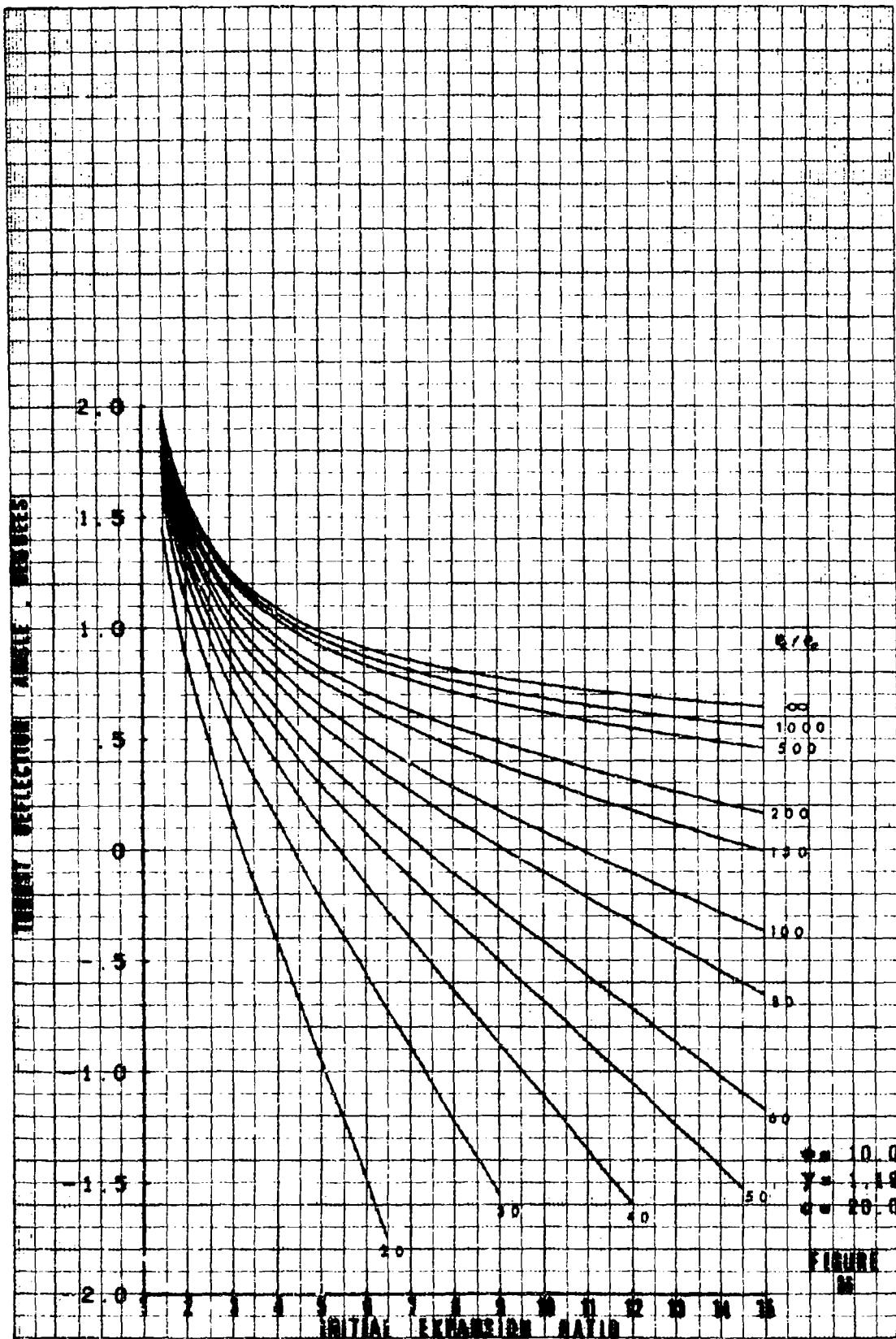
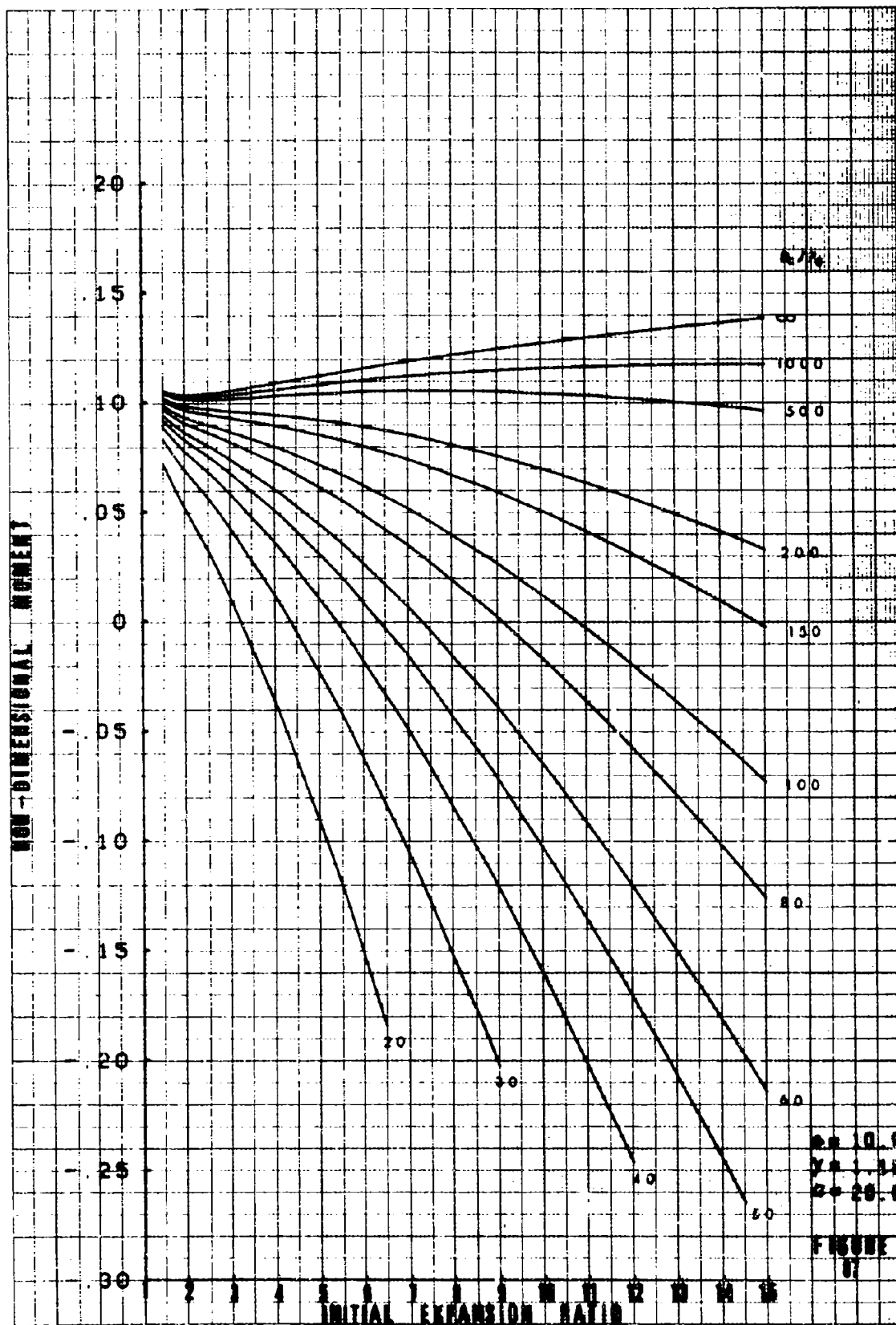


FIGURE 26



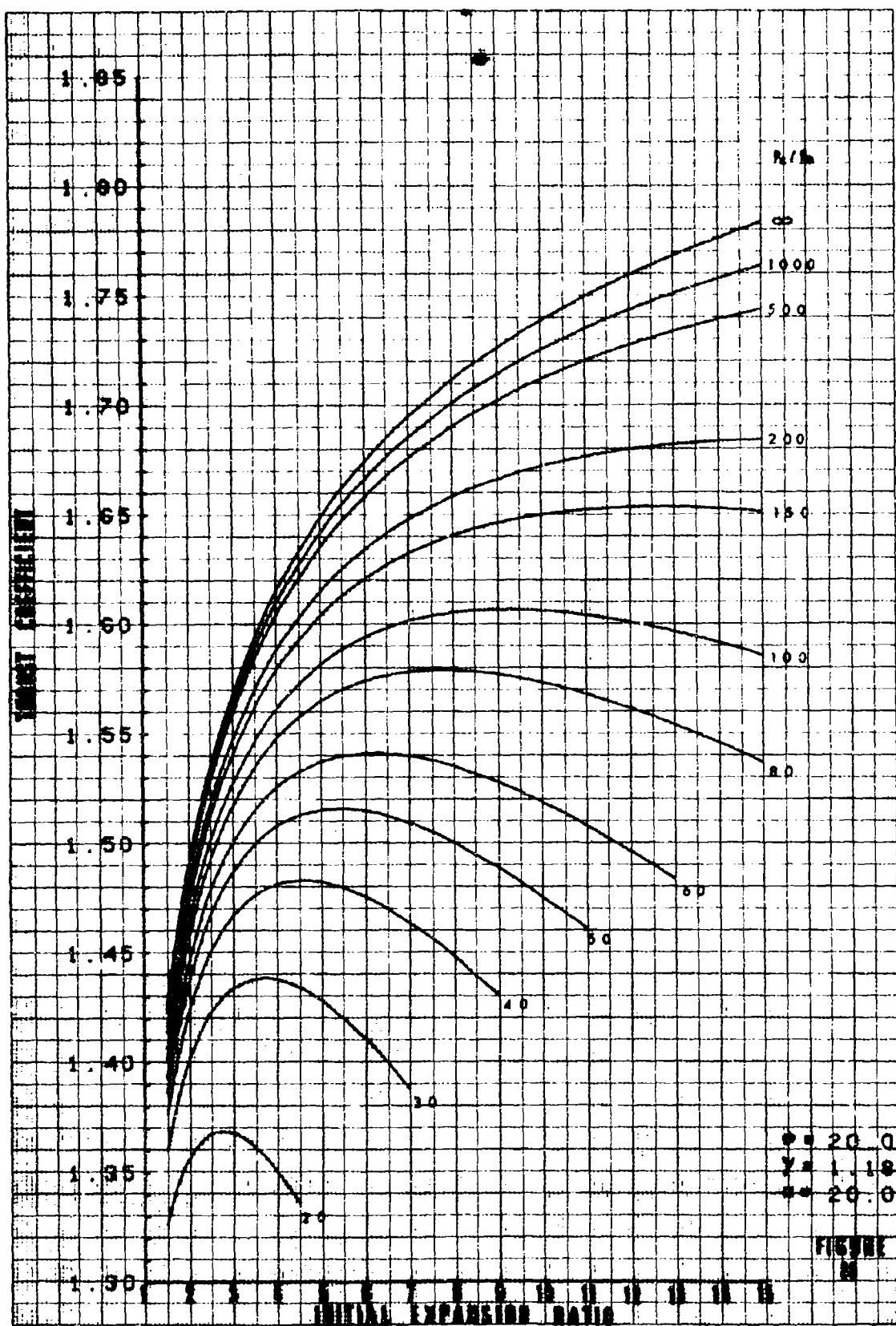
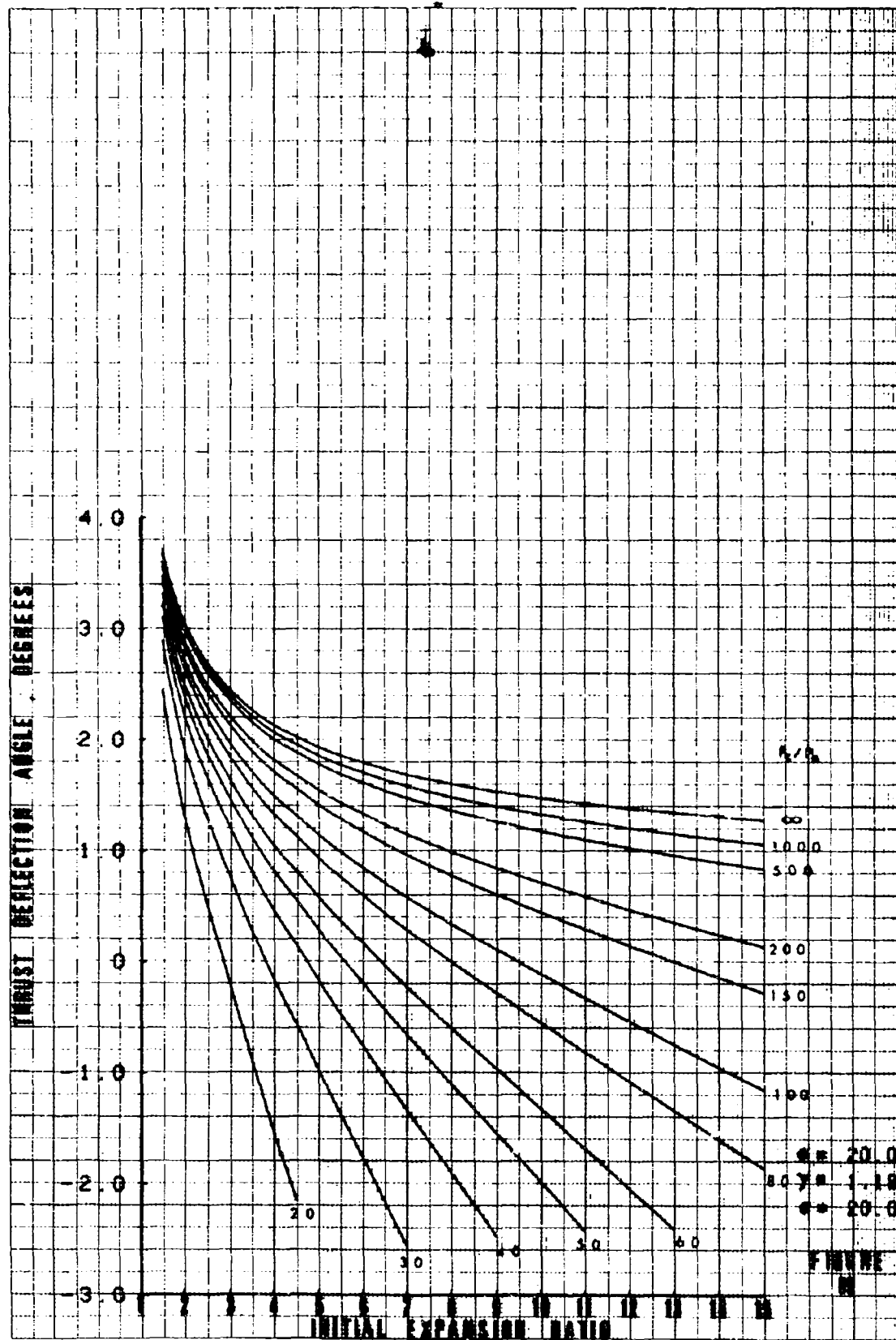
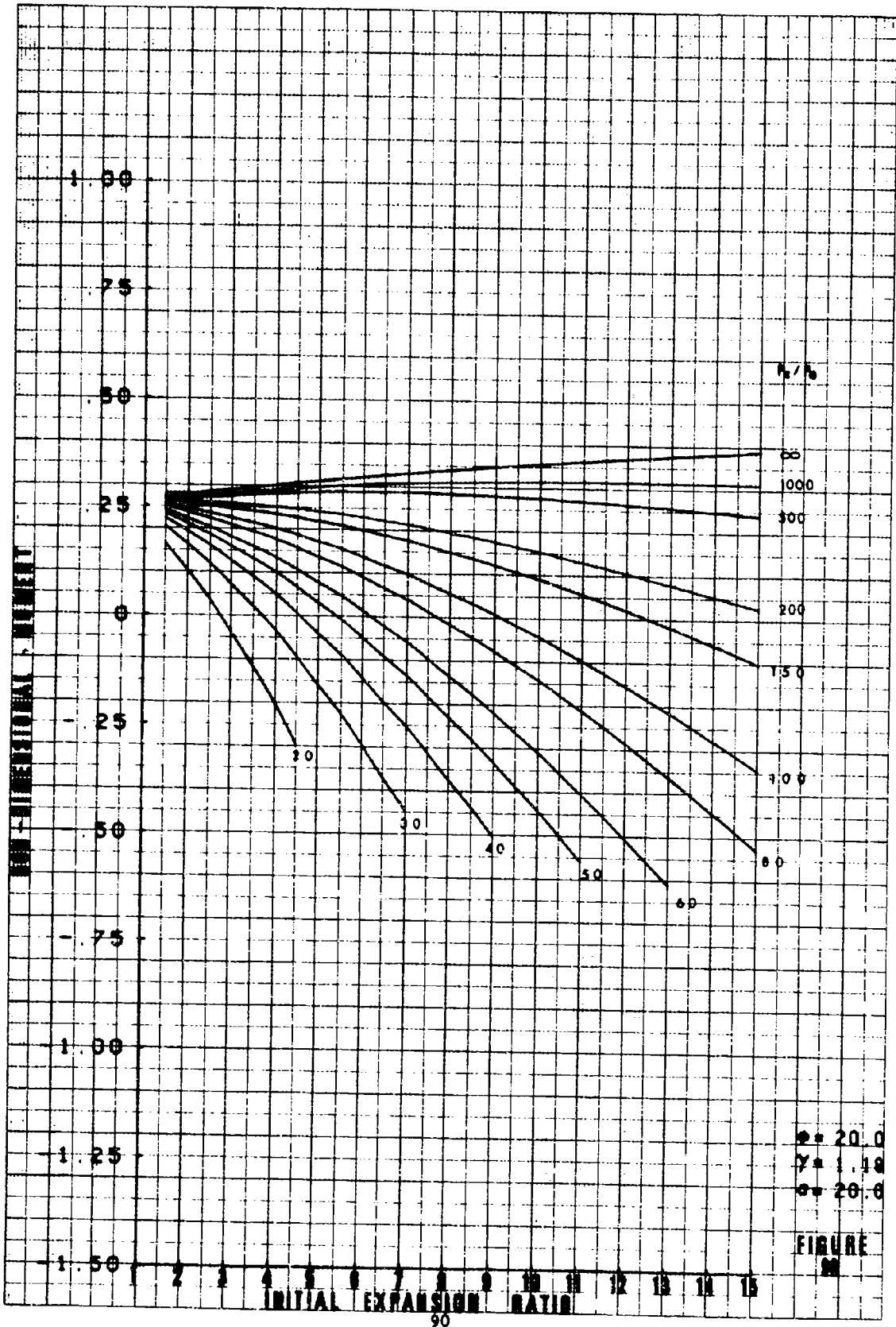


FIGURE 88





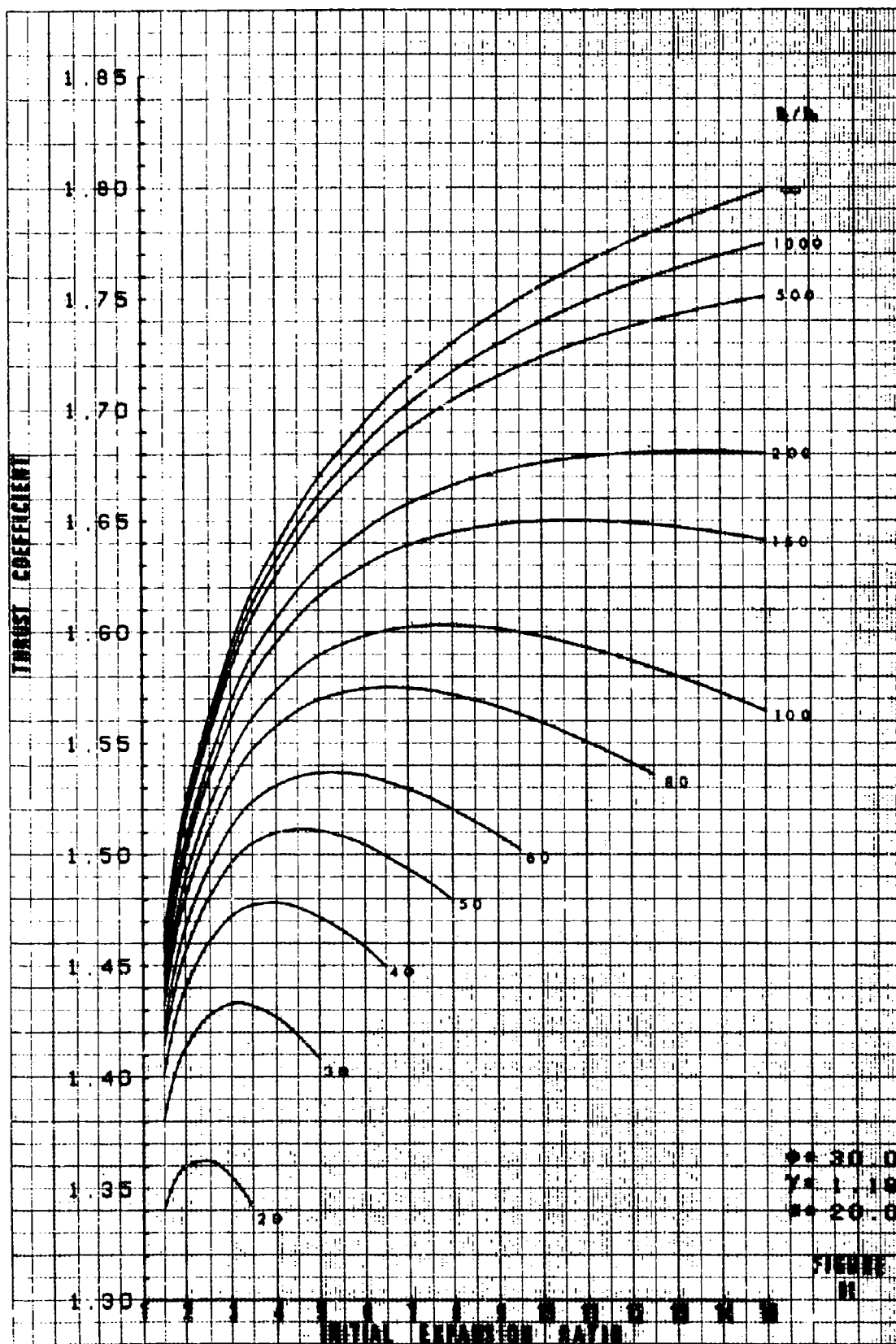
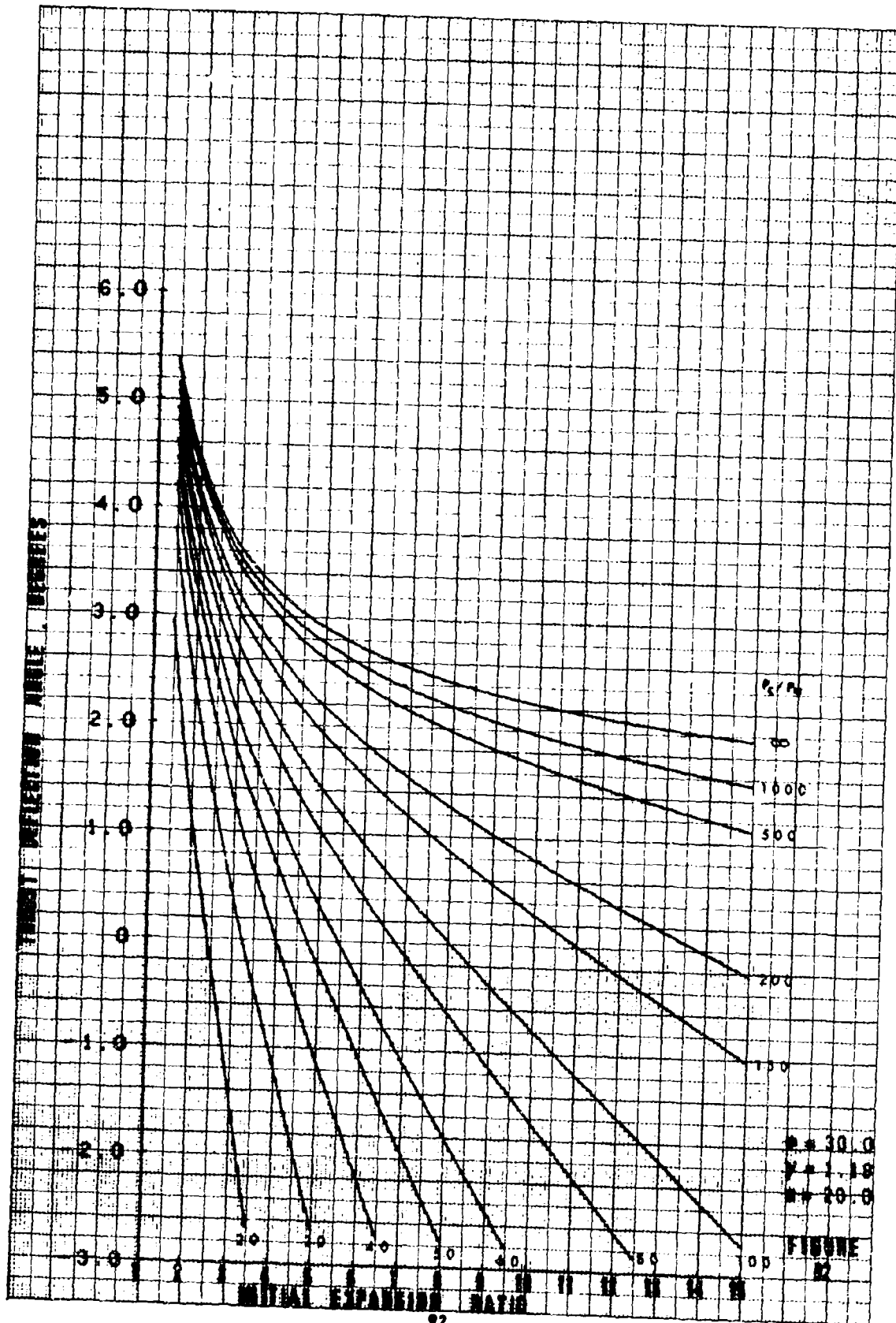
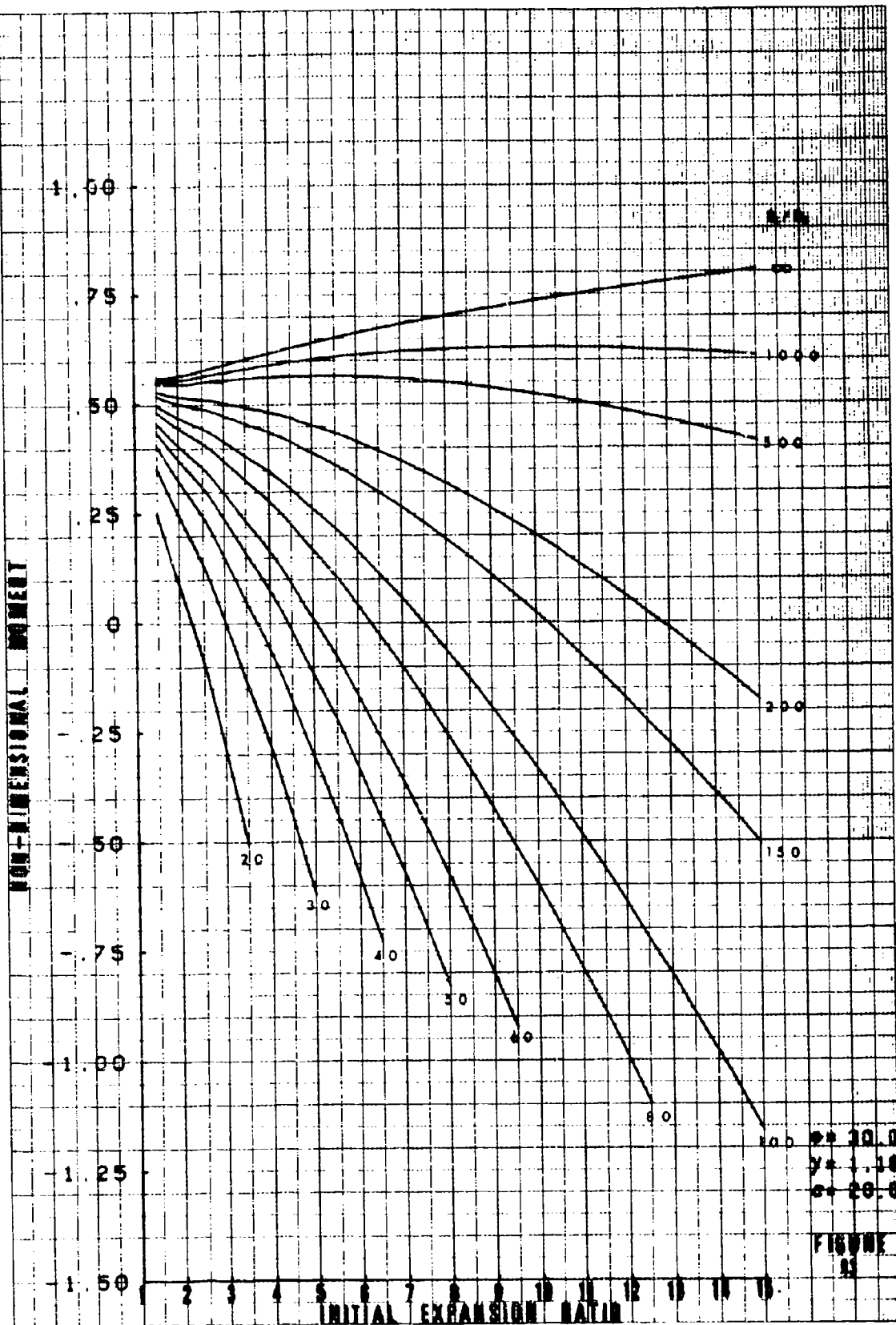
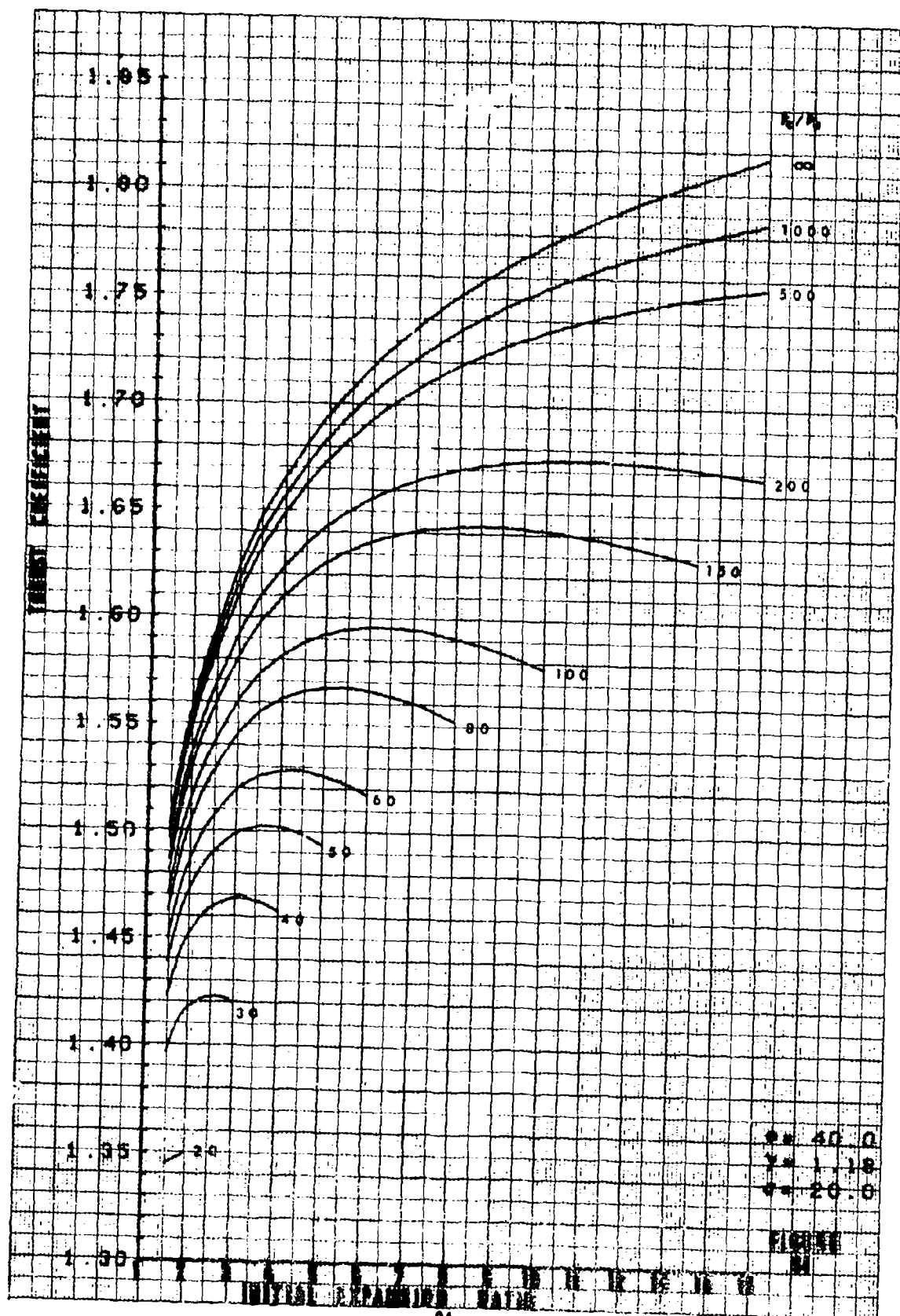
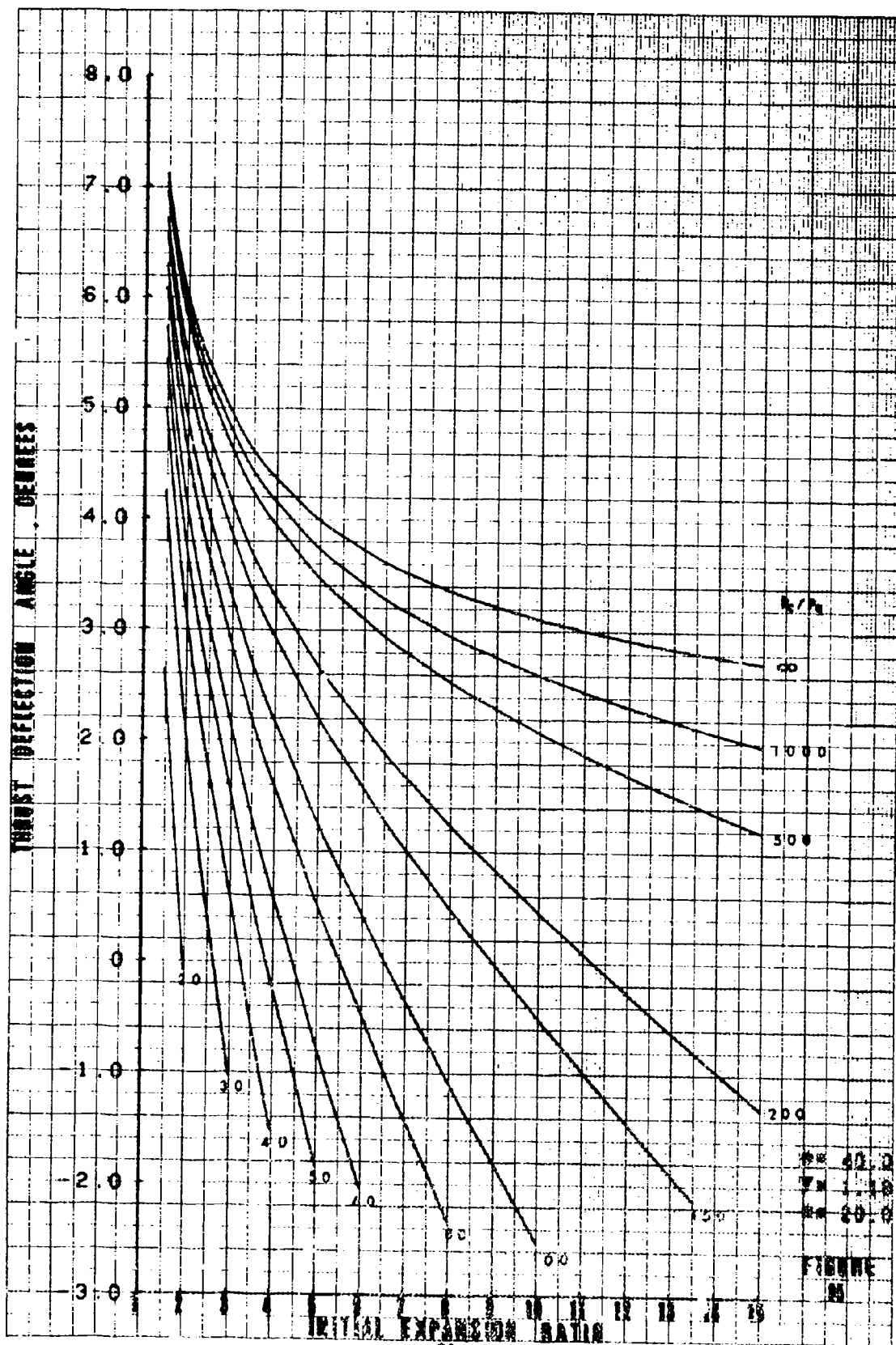


FIGURE
11

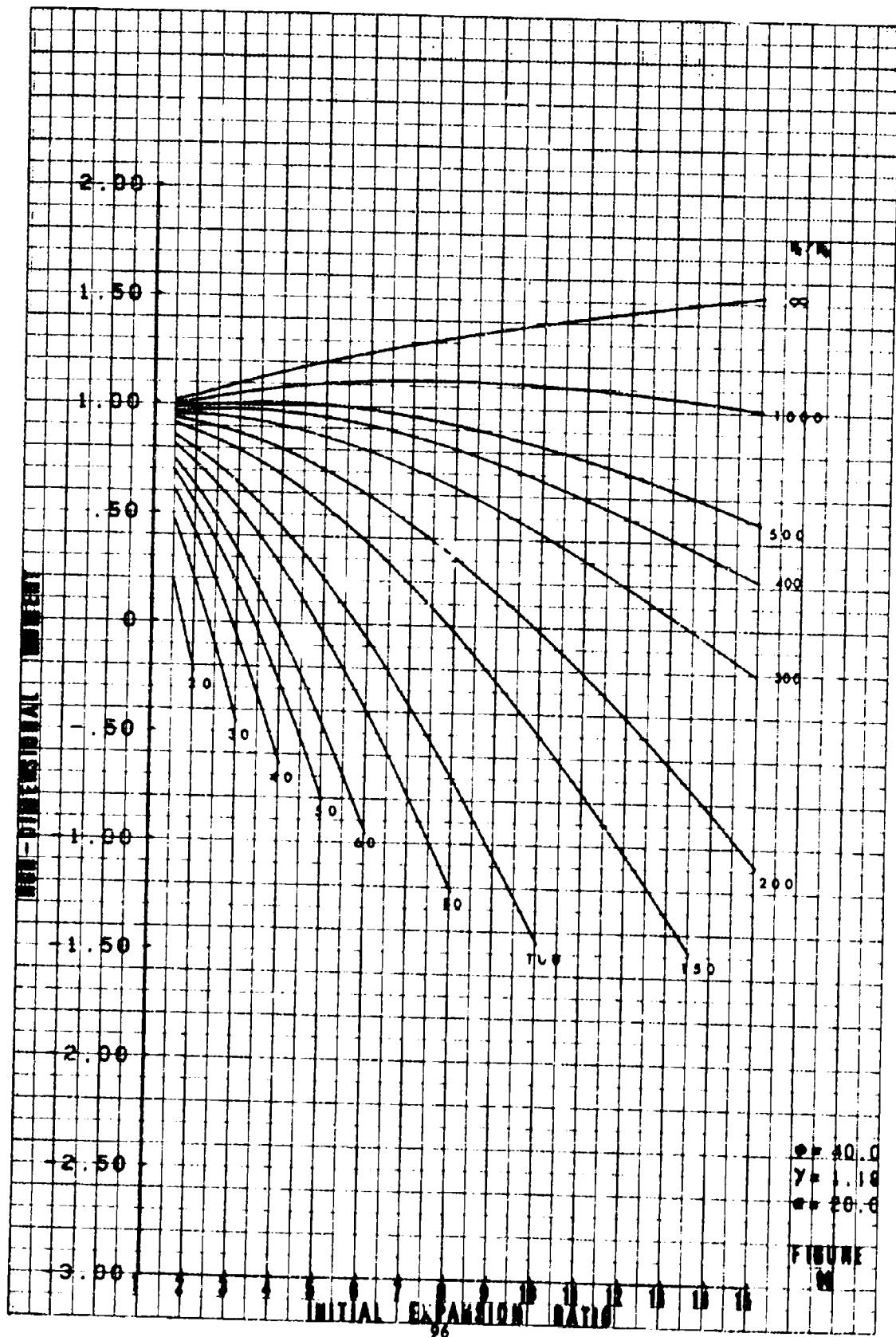


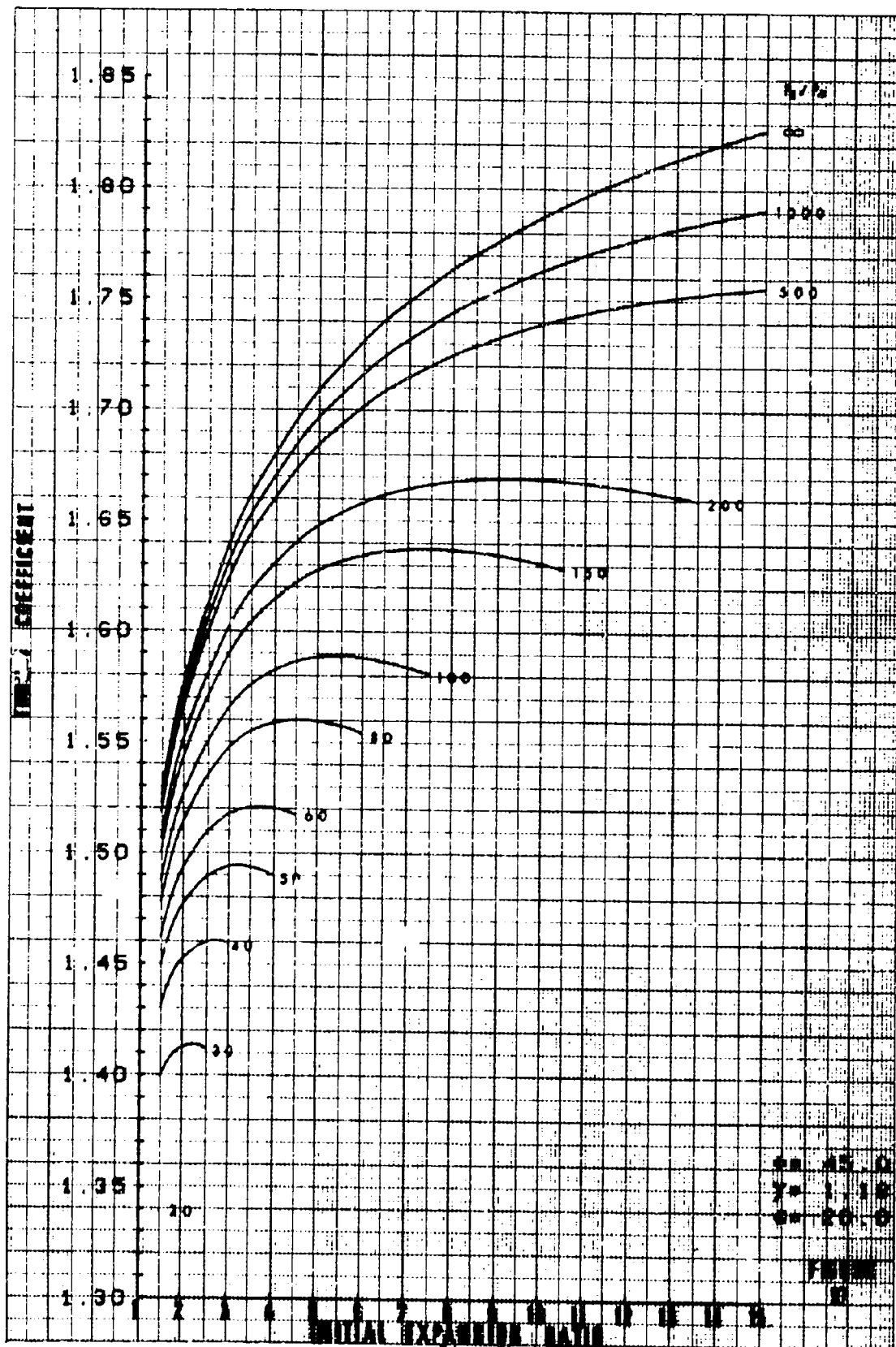


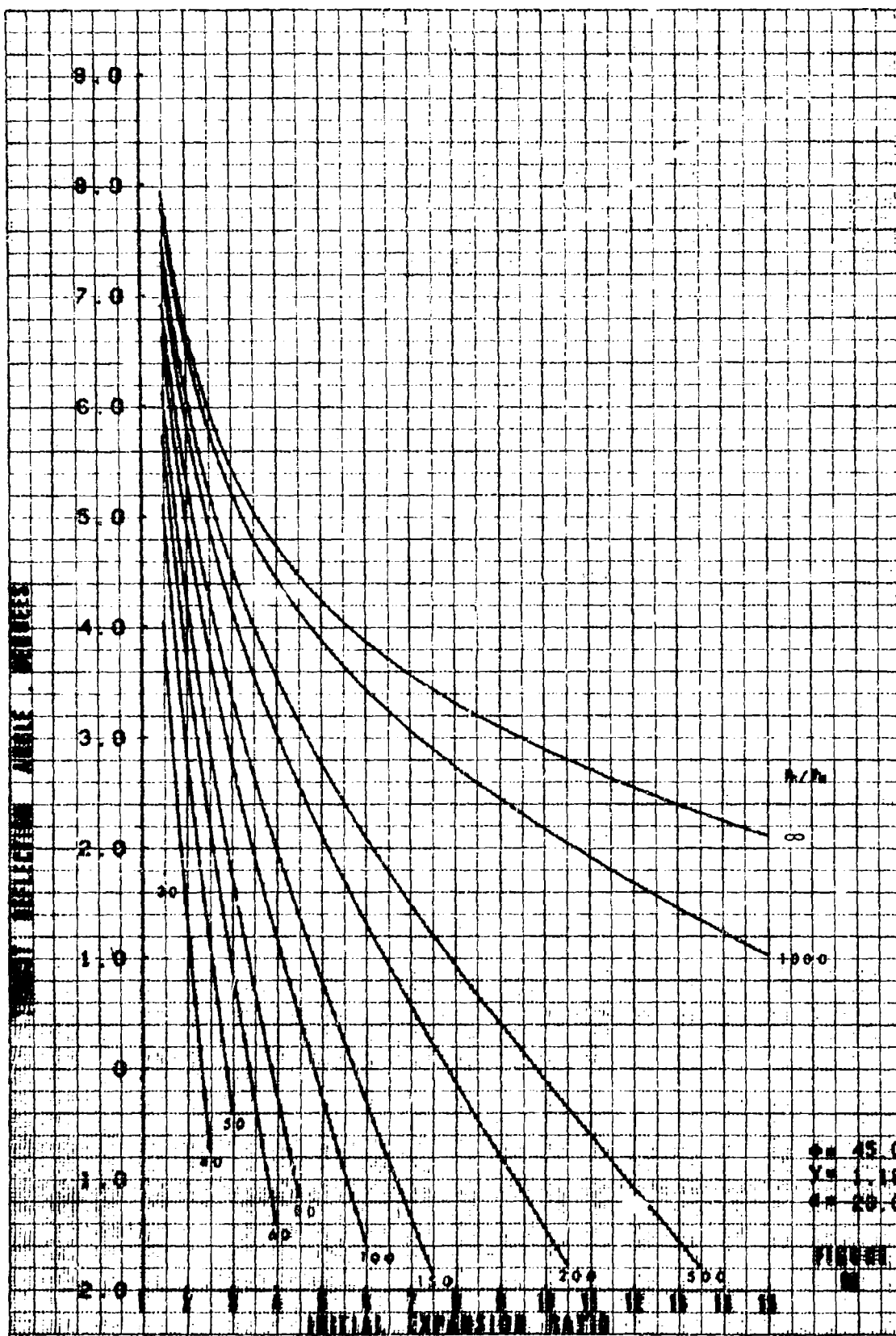


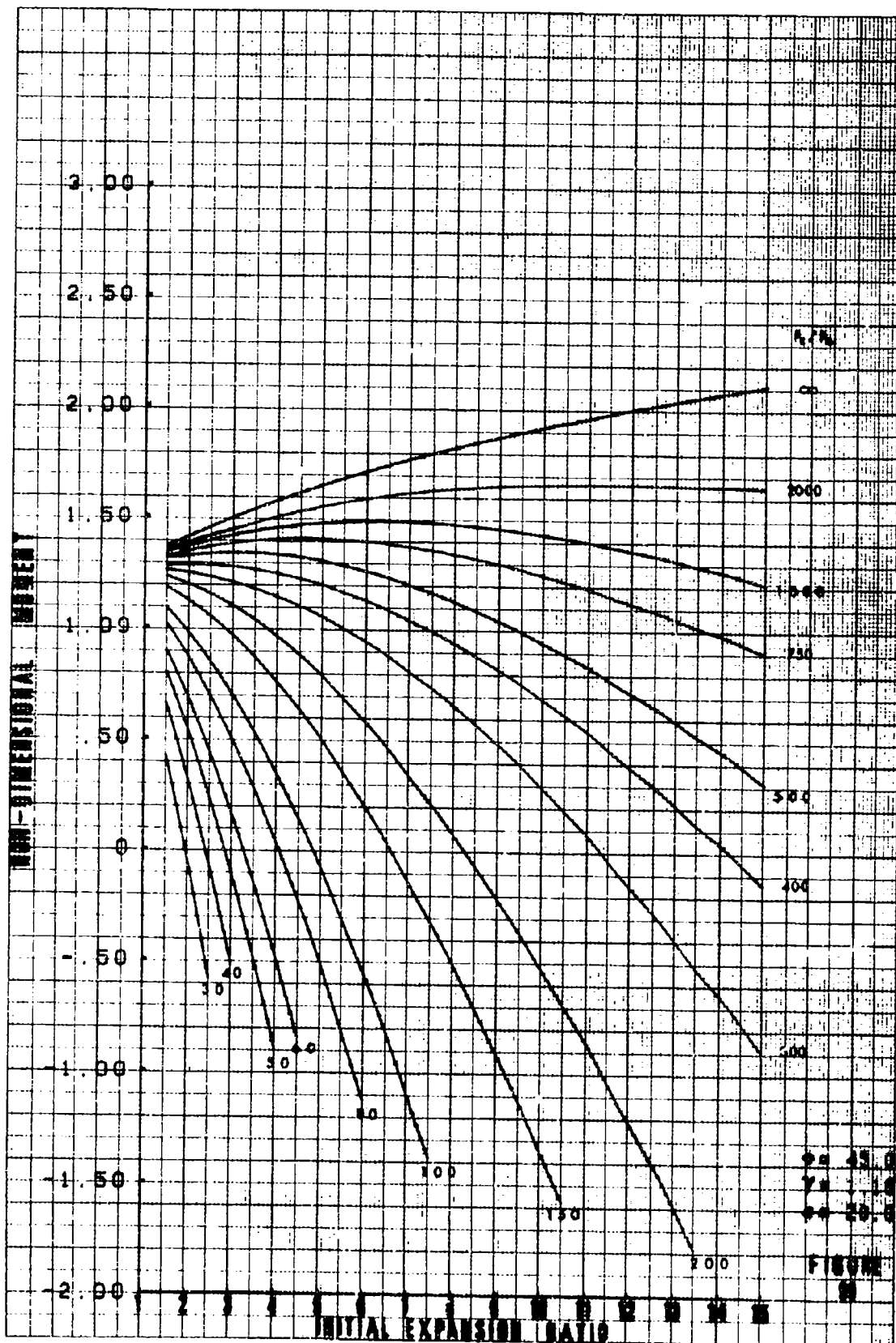


$\gamma = 1.4$
 $M = 20.0$
 FIGURE 24

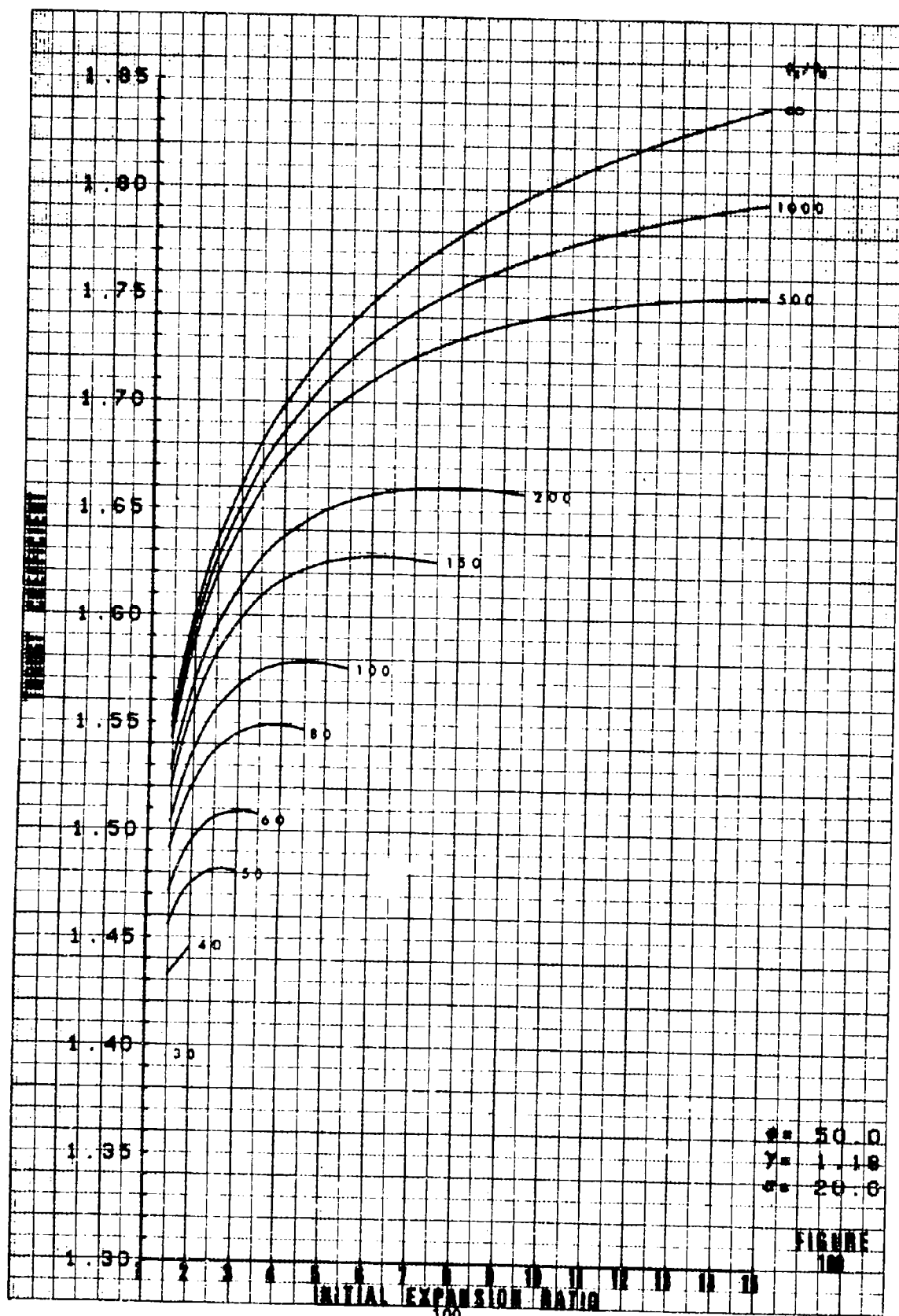








FIGURE



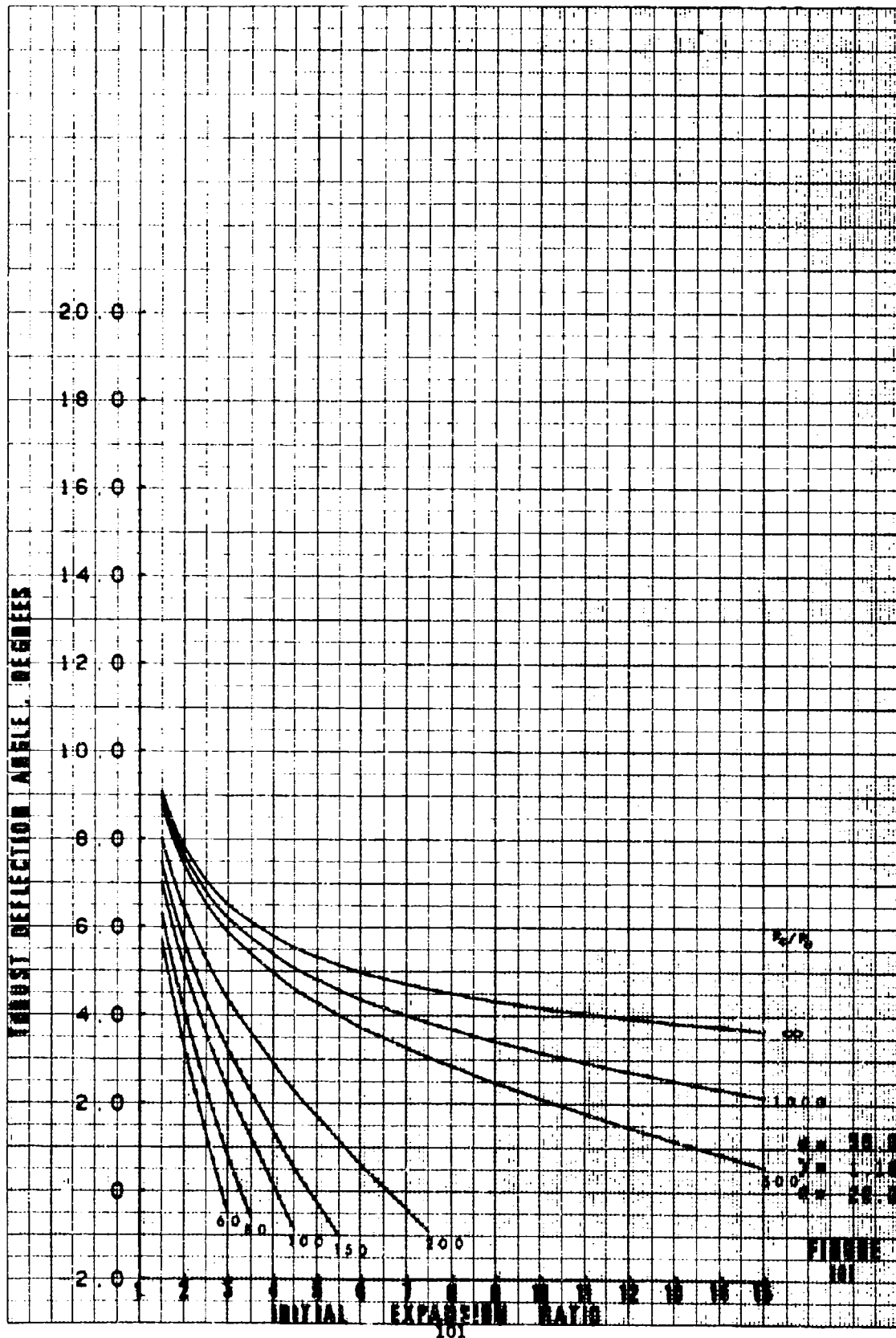
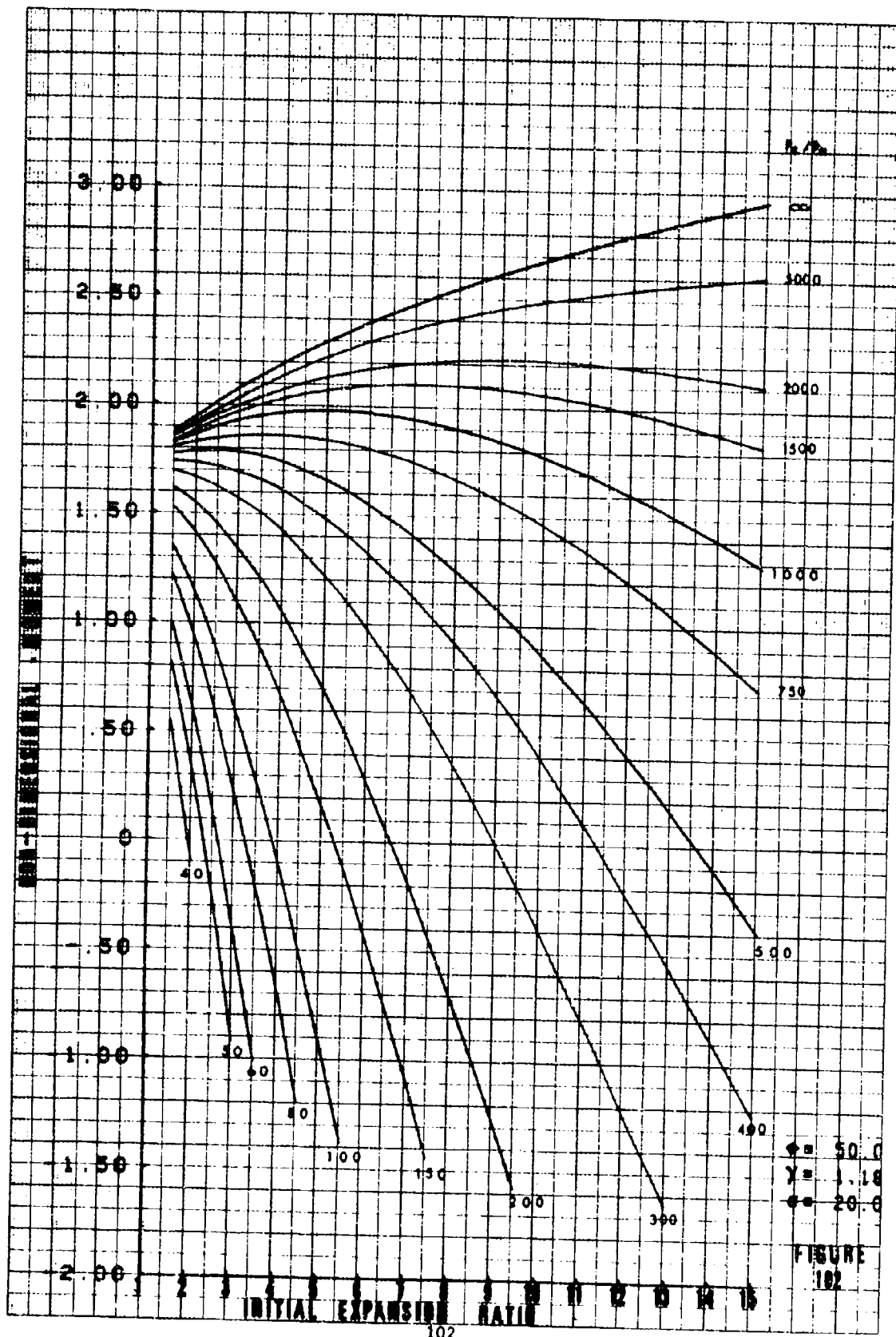
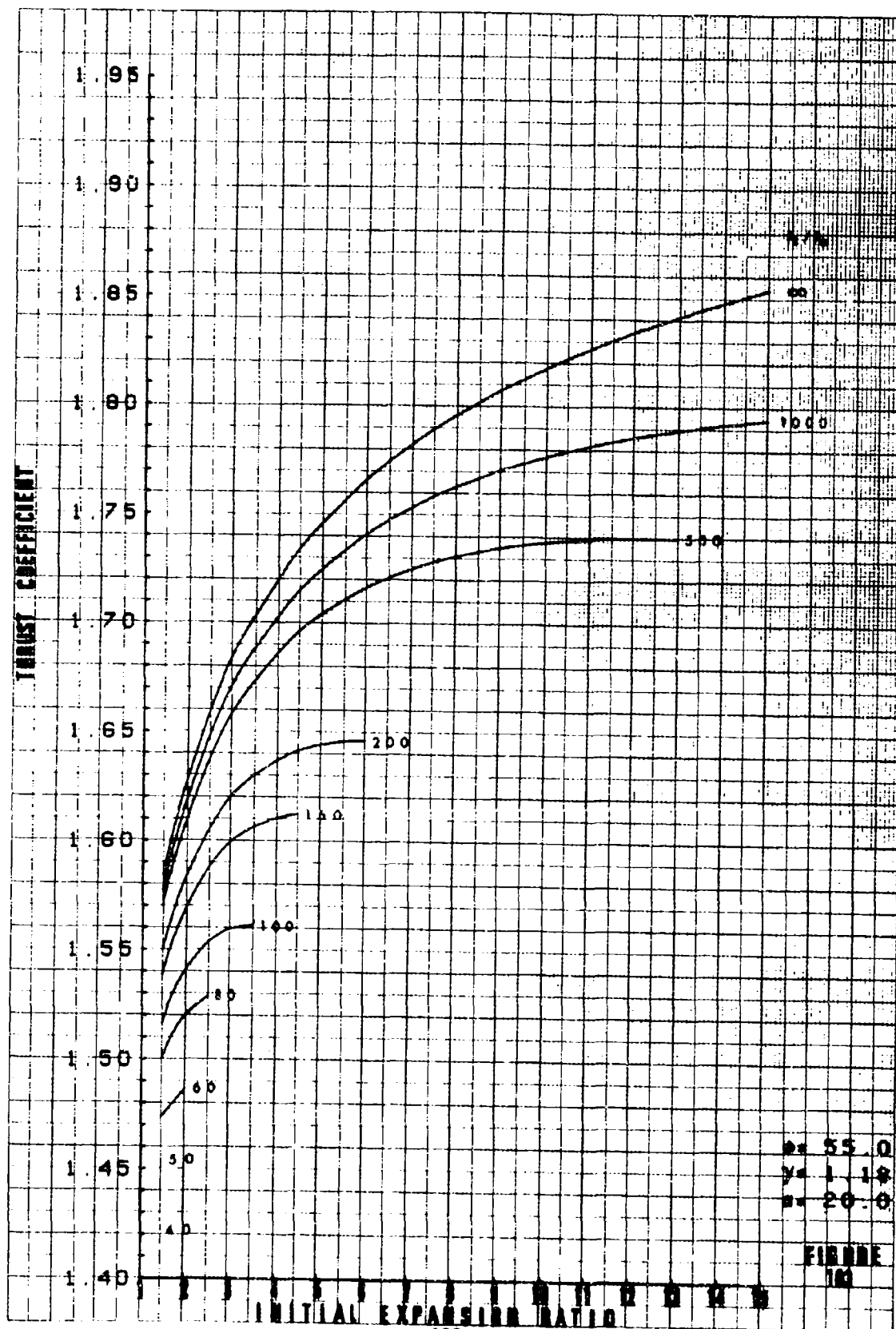
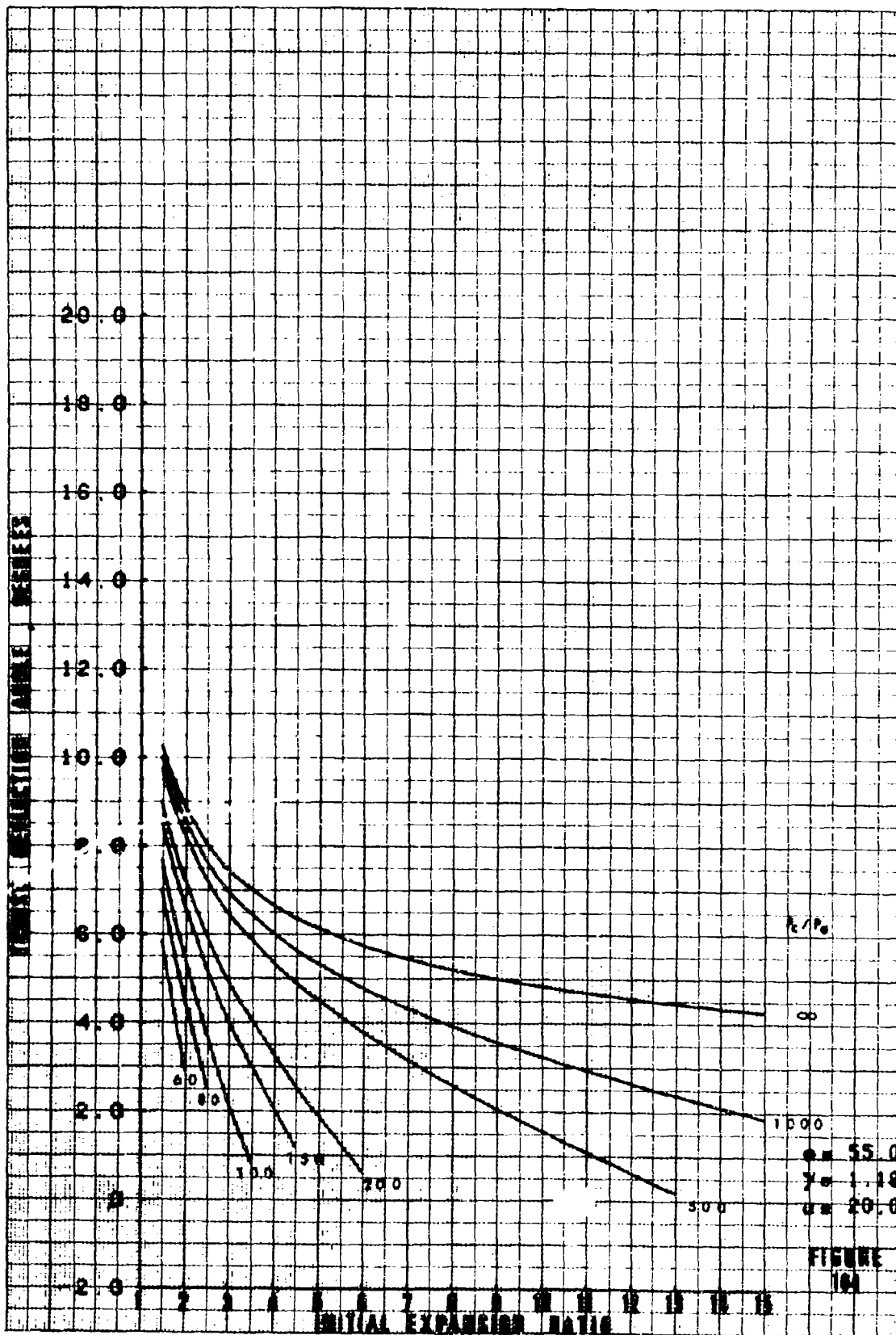
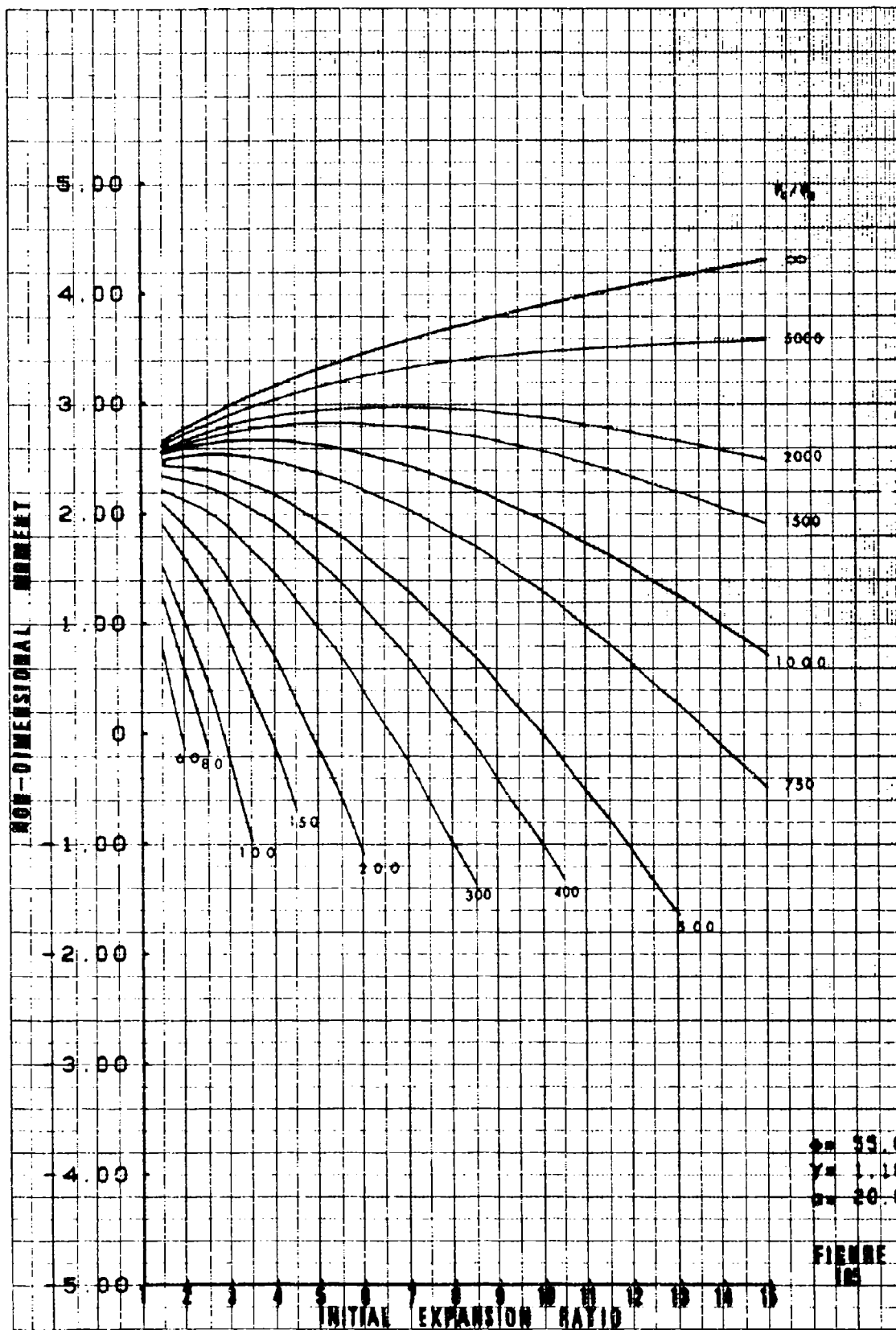


FIGURE 101









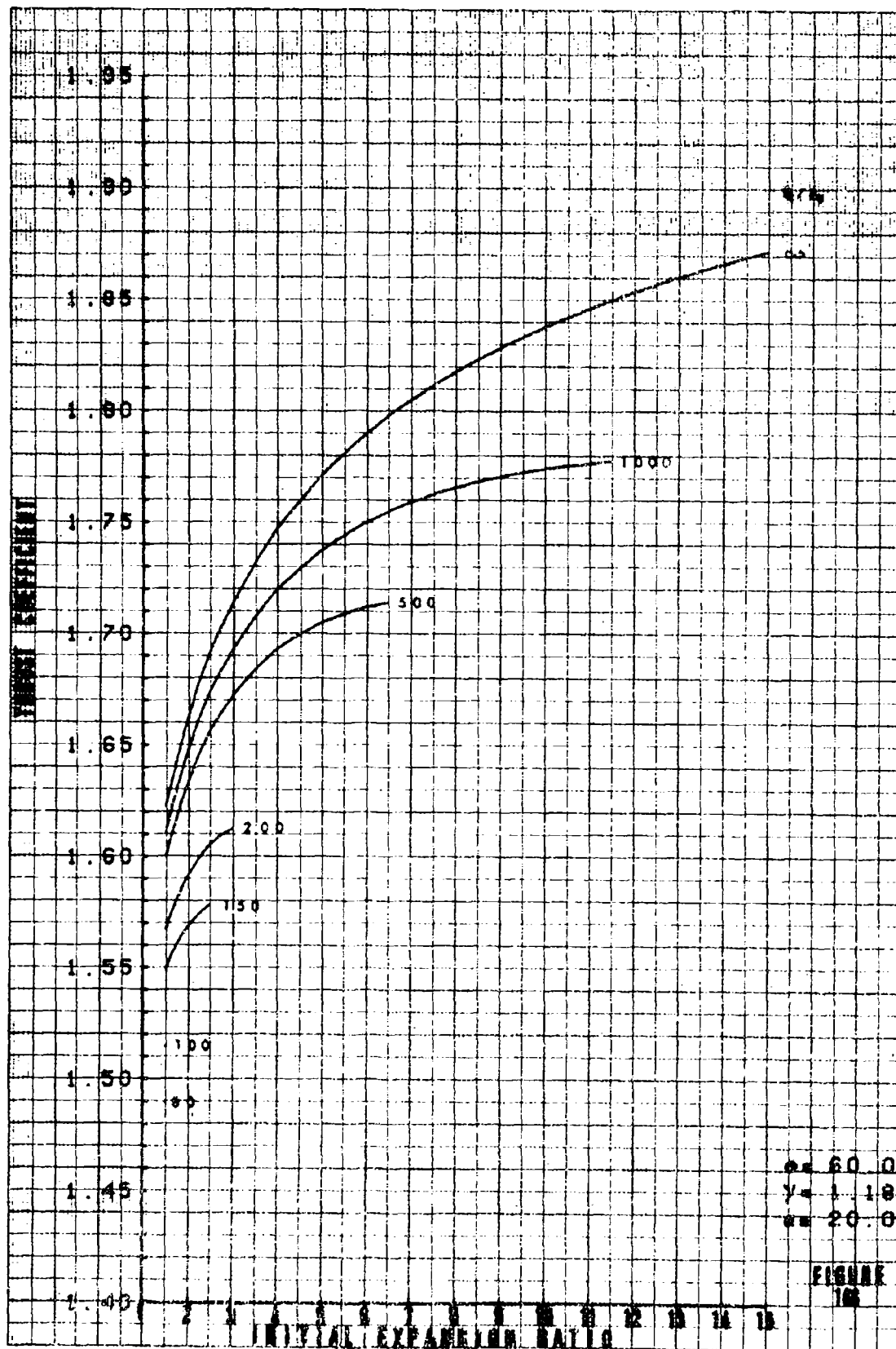
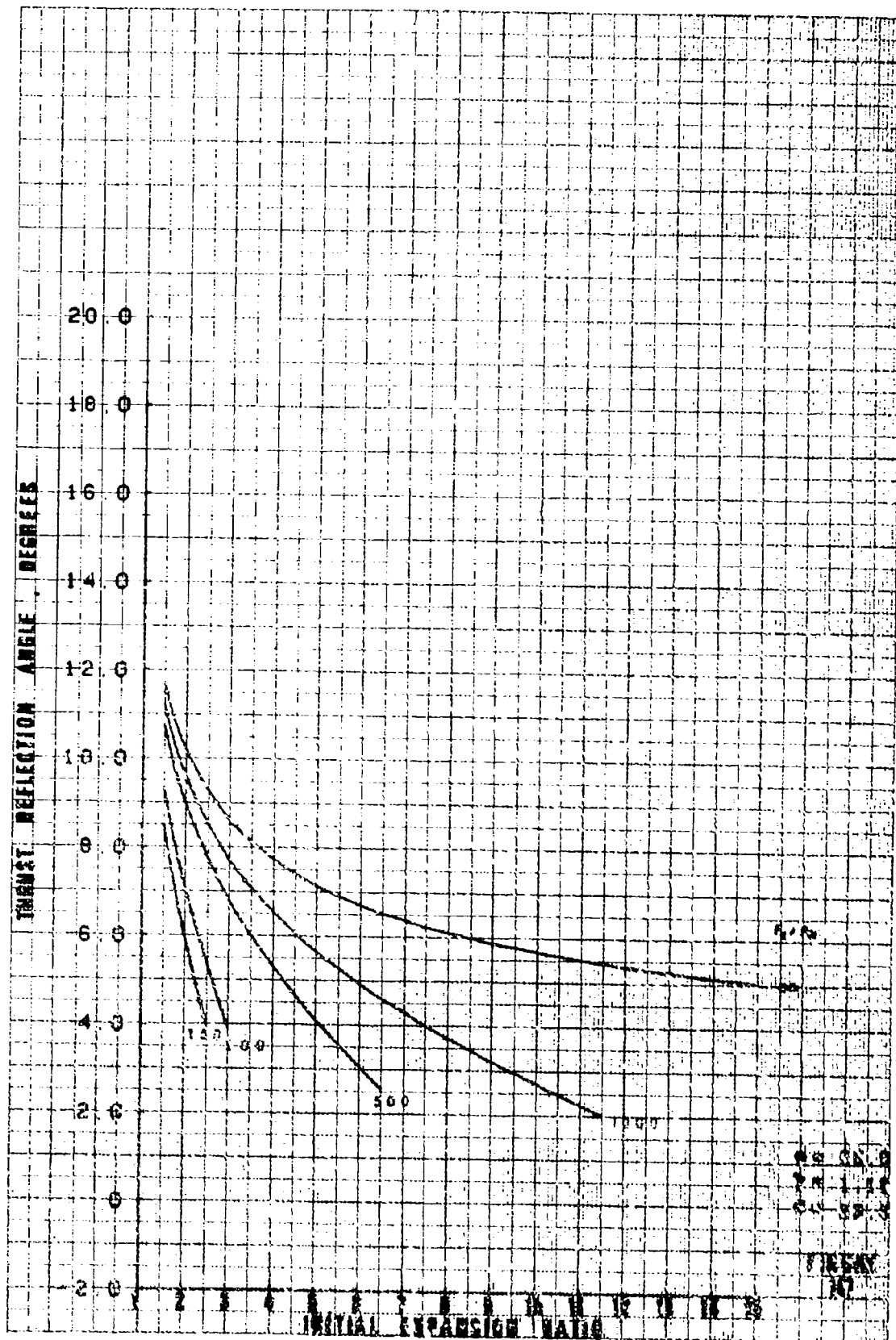
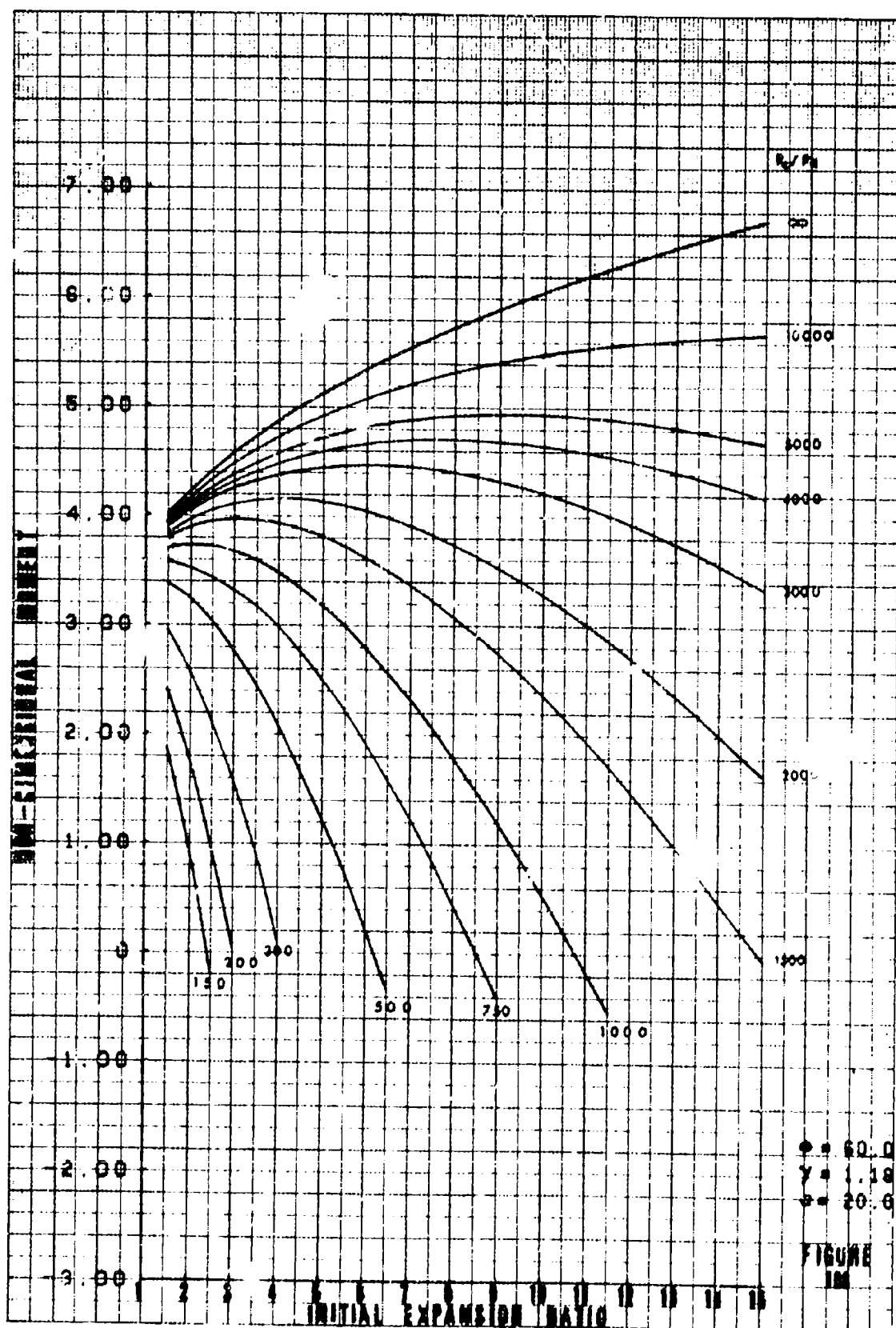
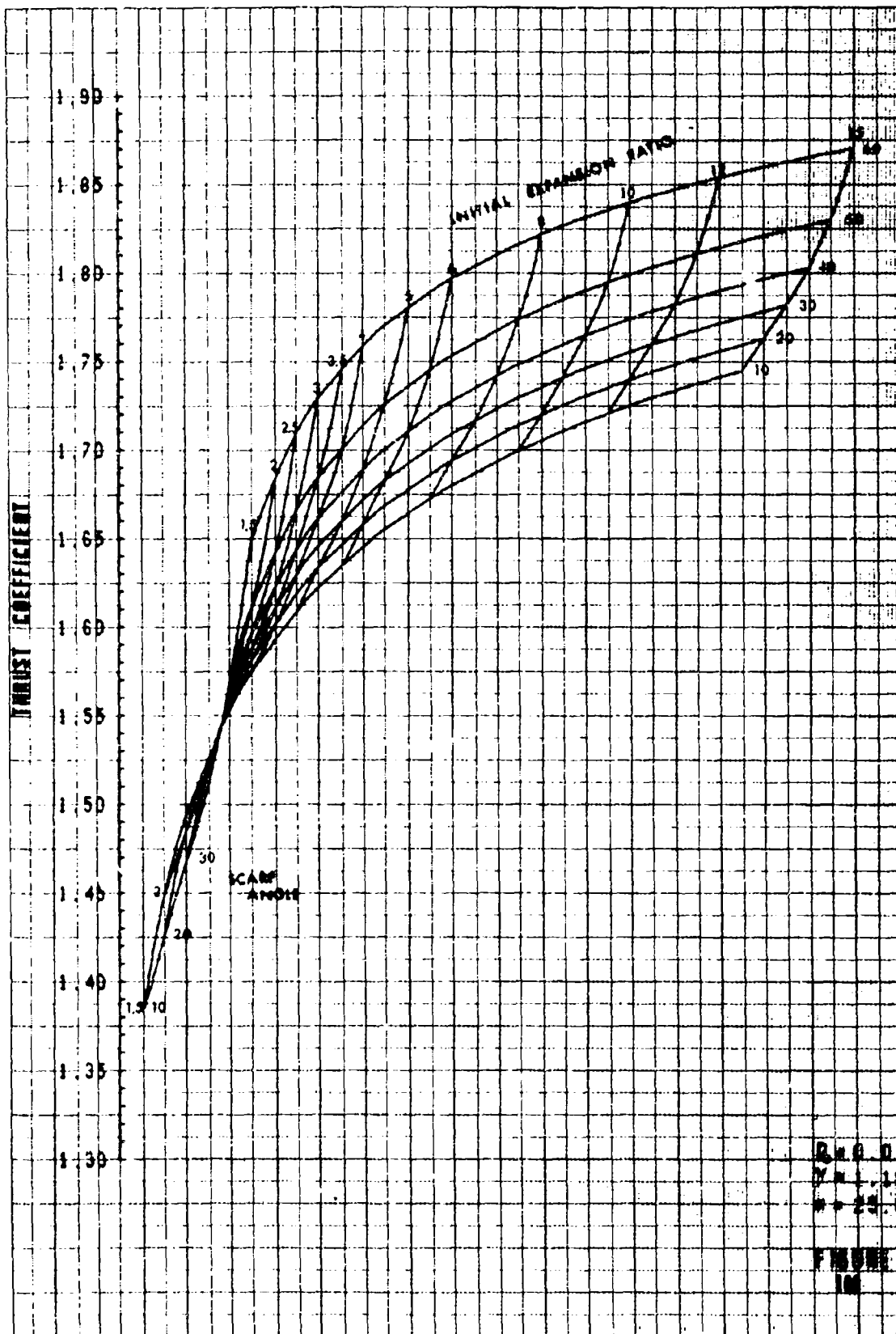
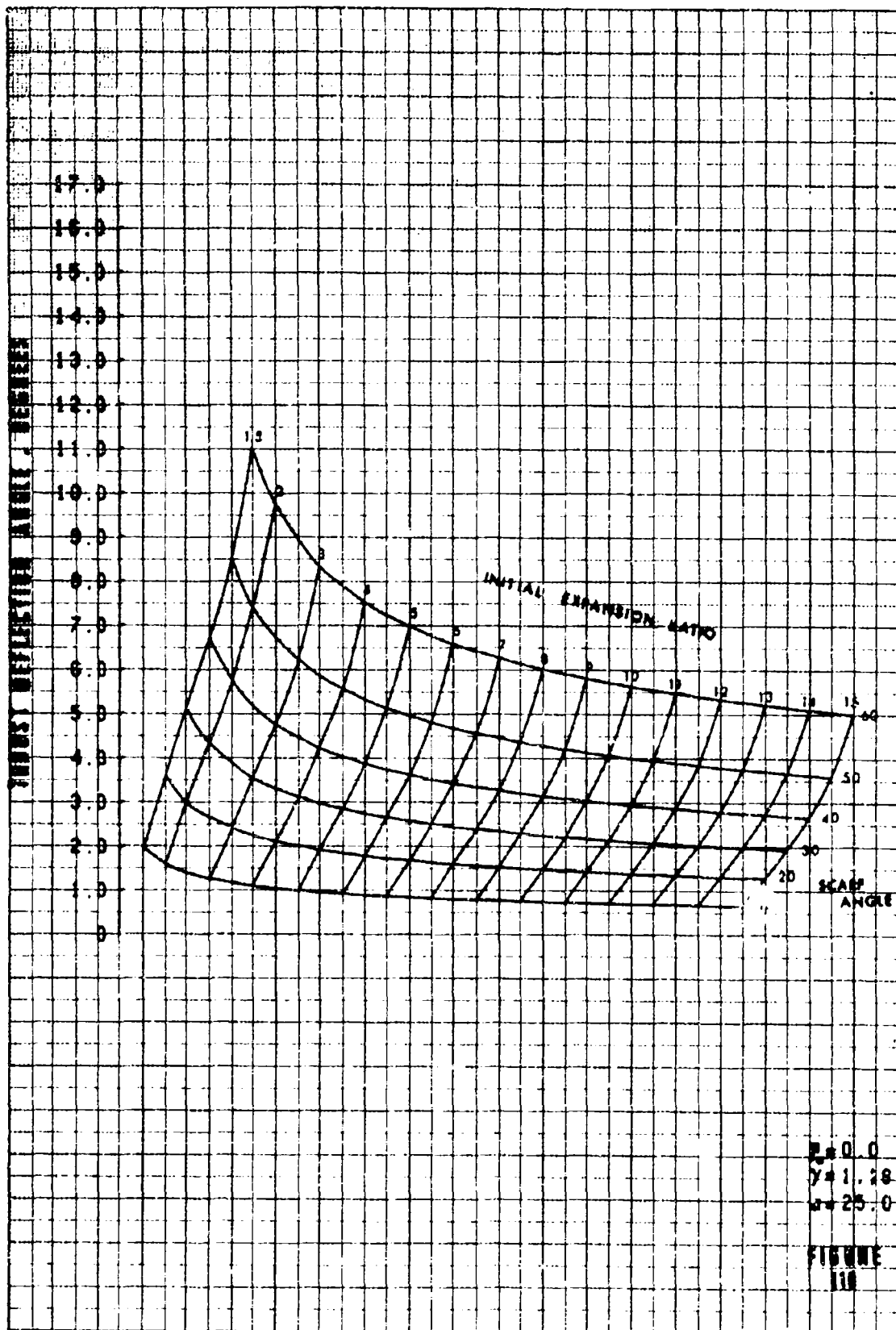


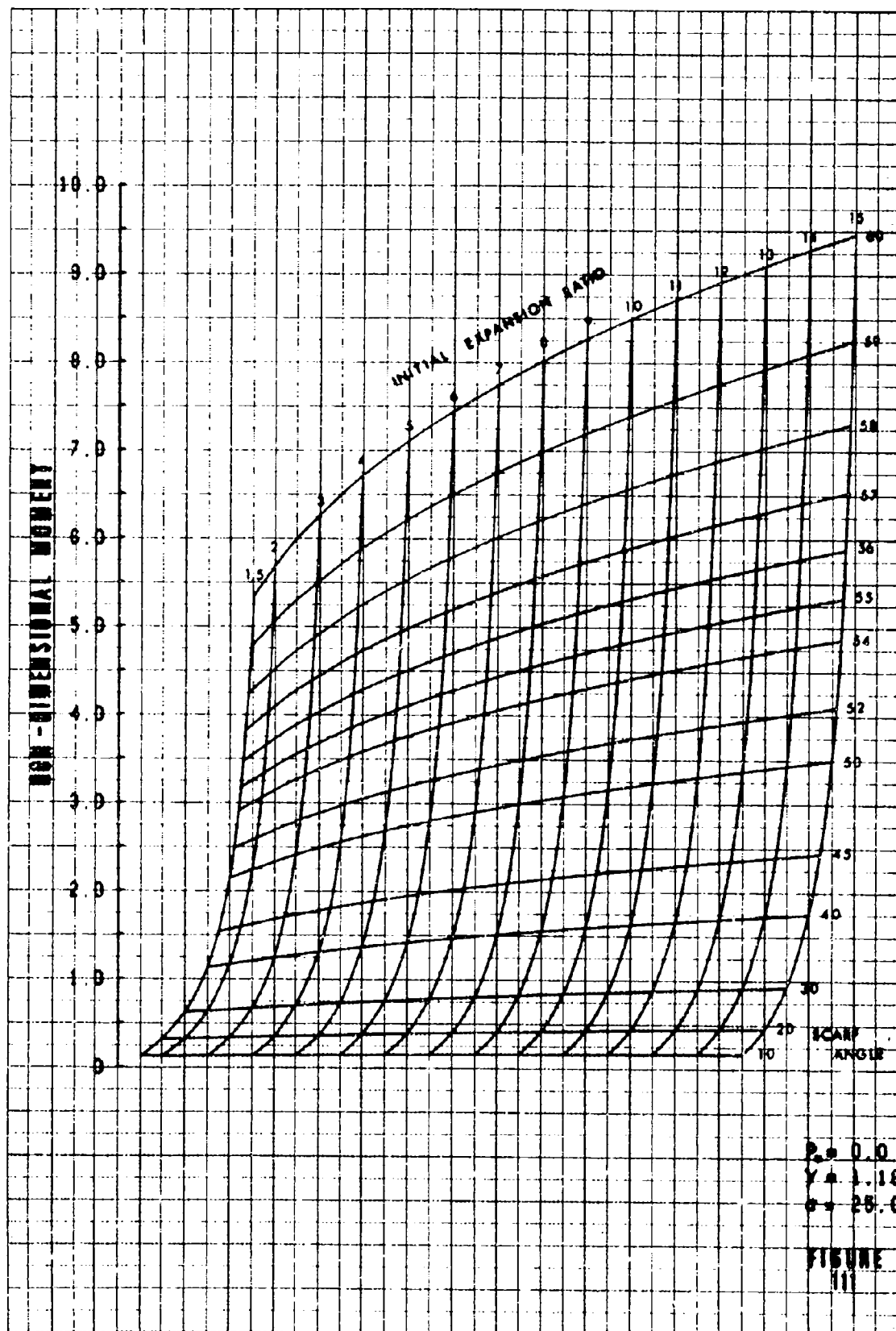
FIGURE 106

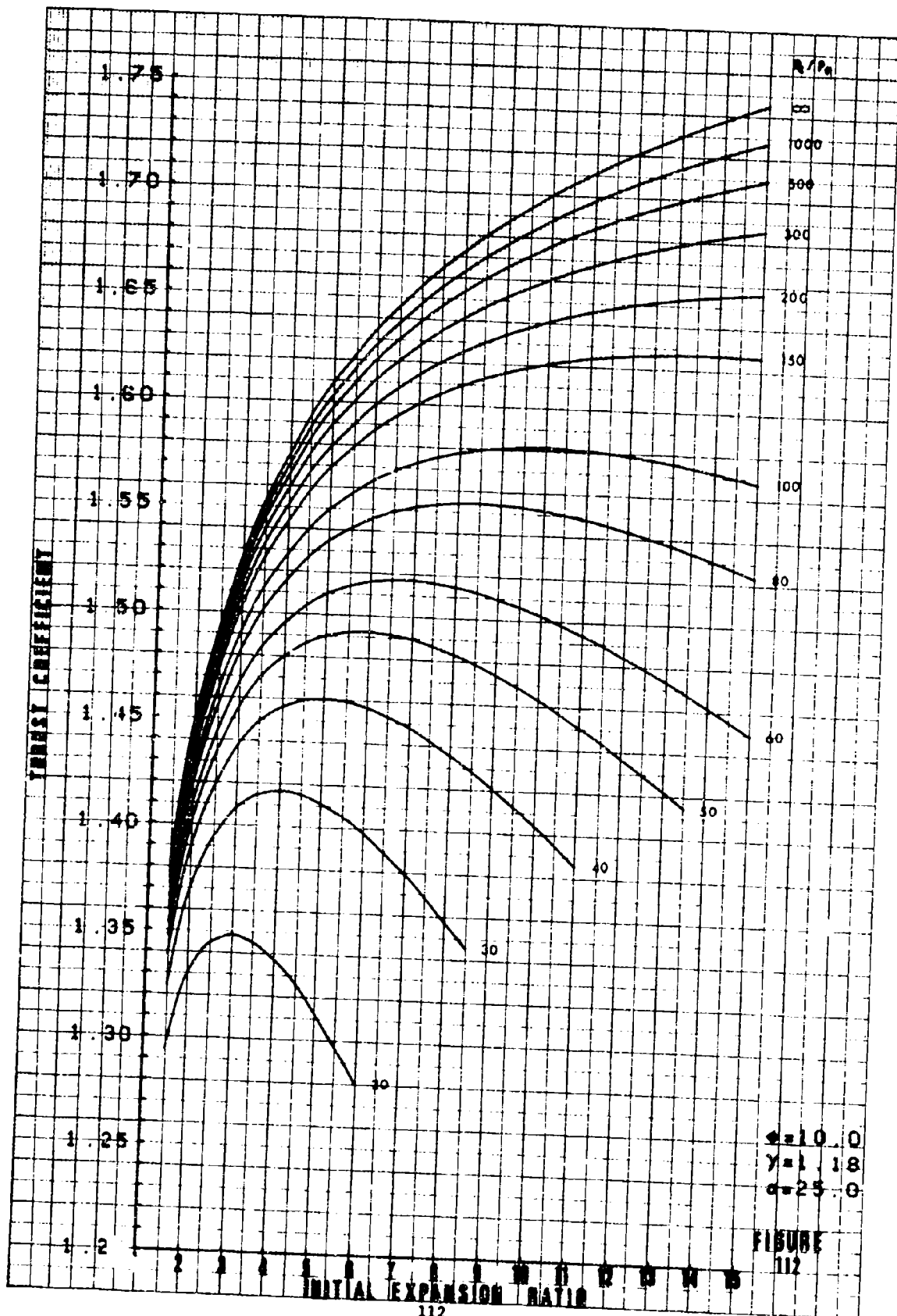


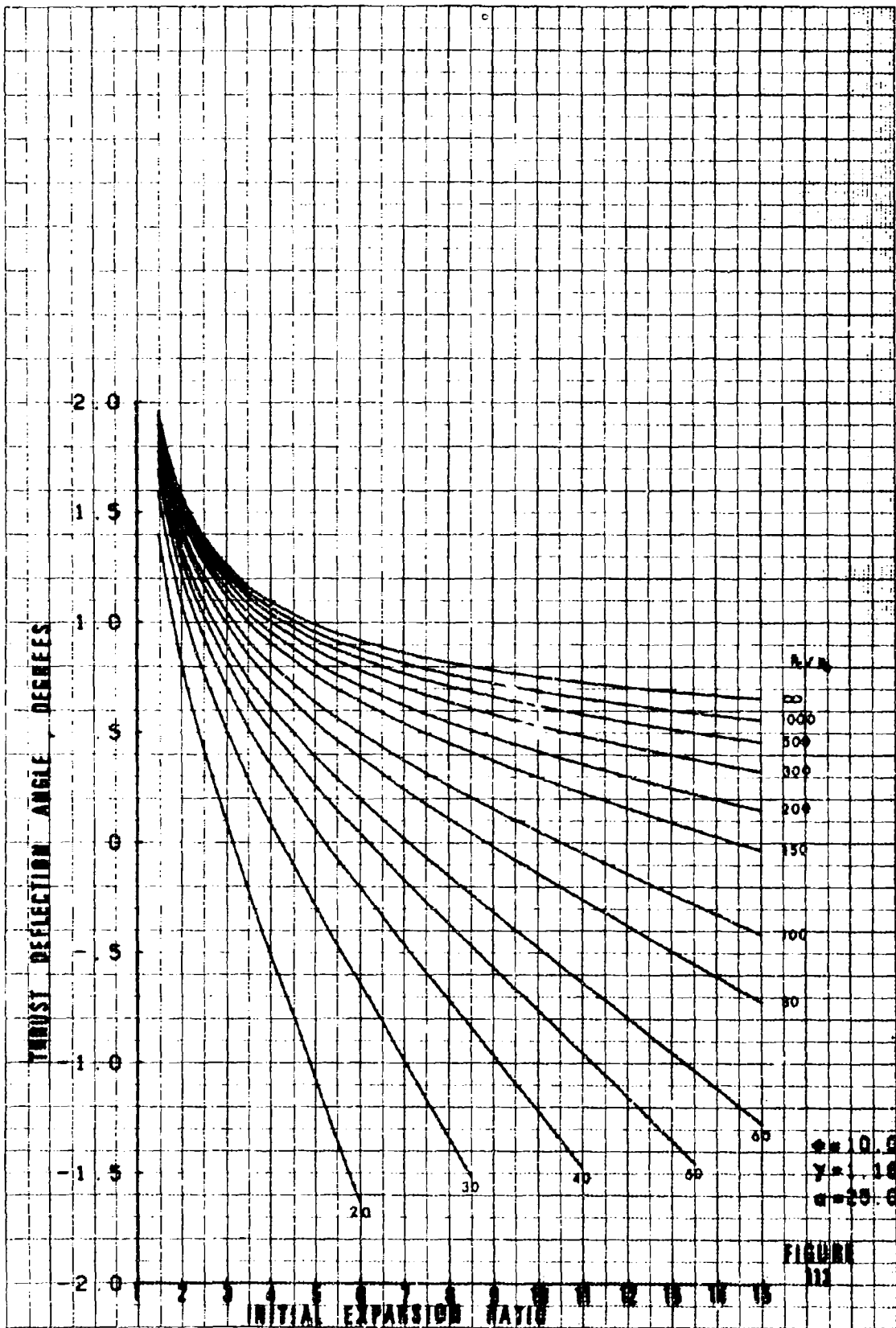


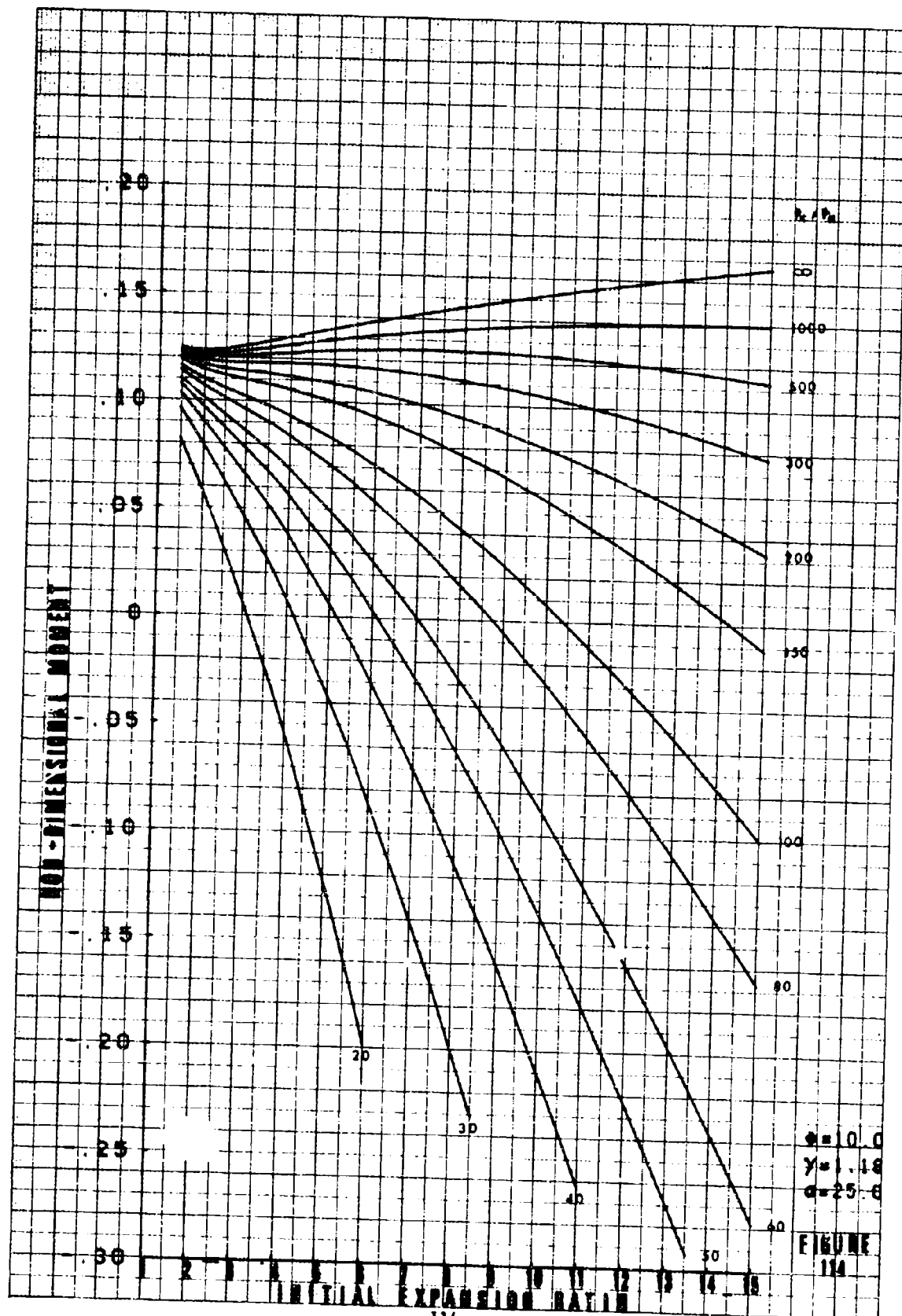


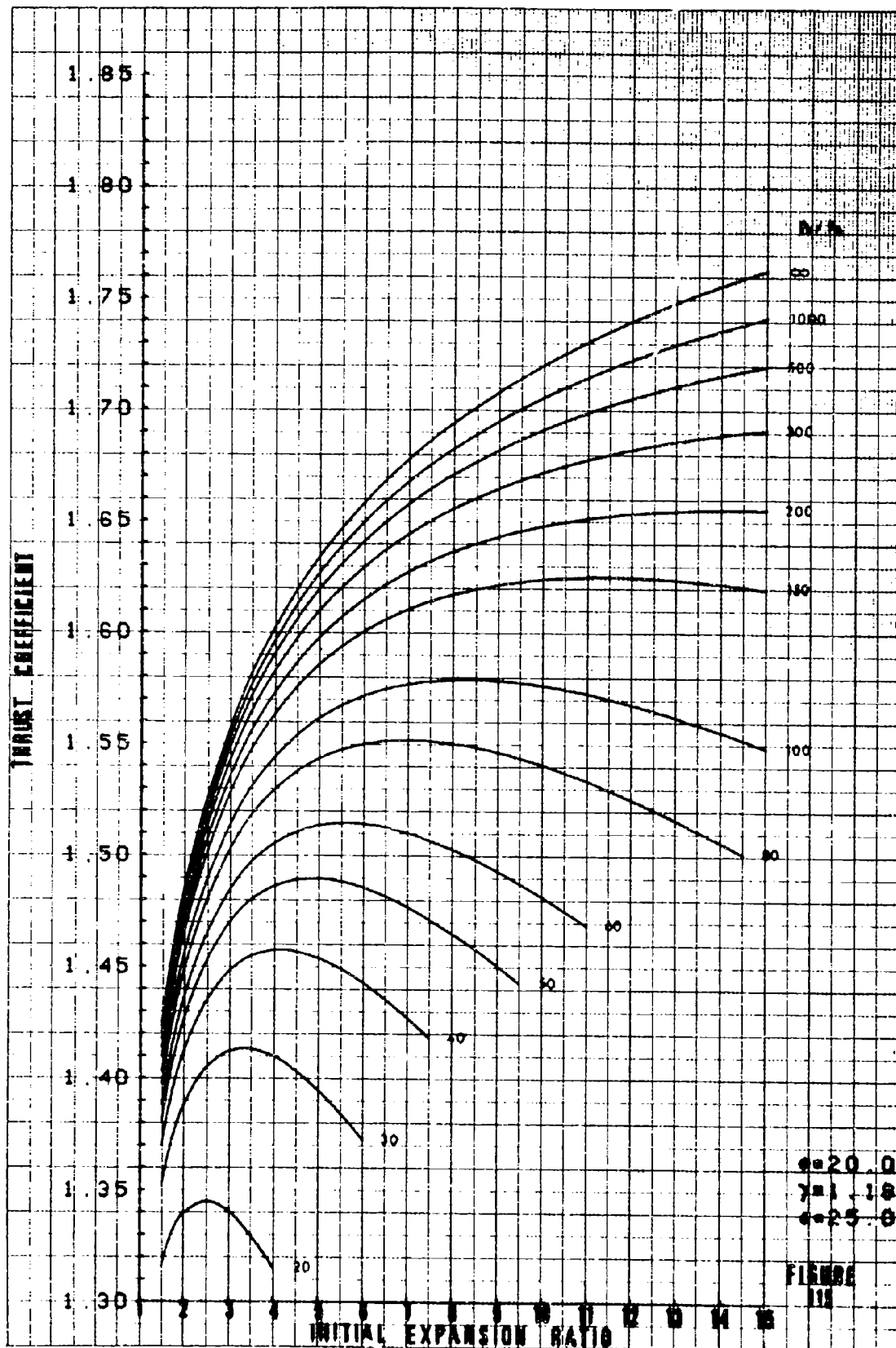












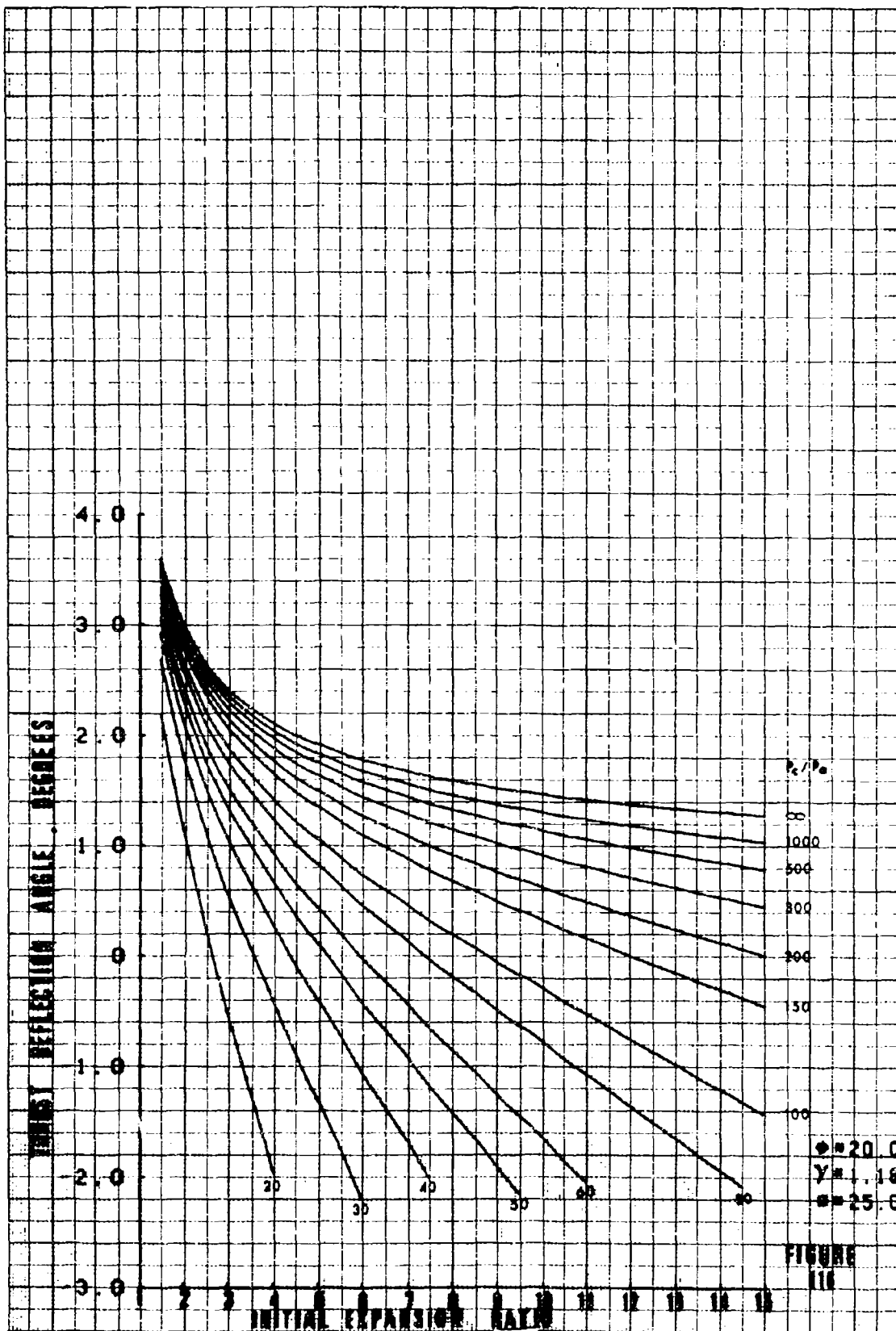


FIGURE 116

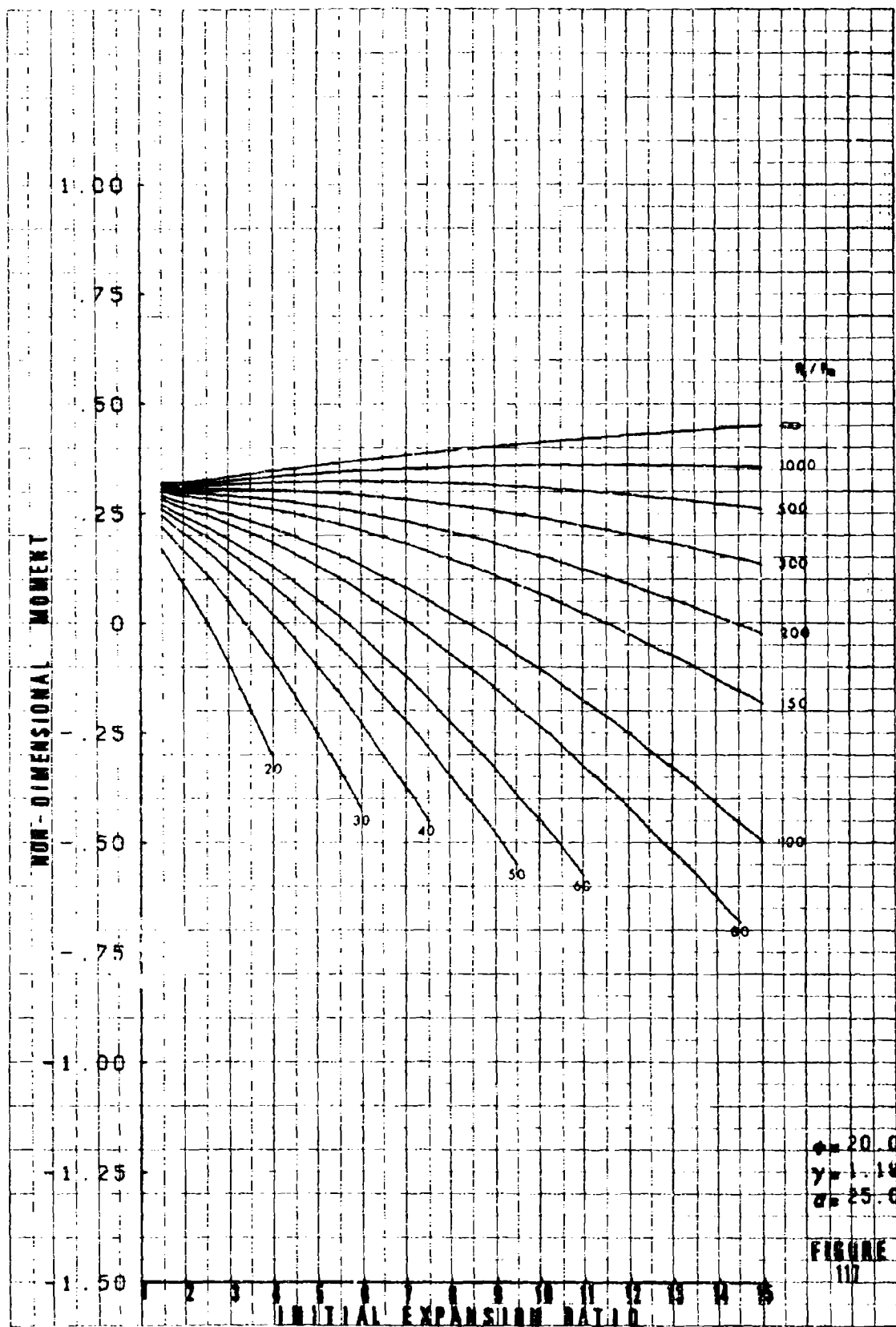
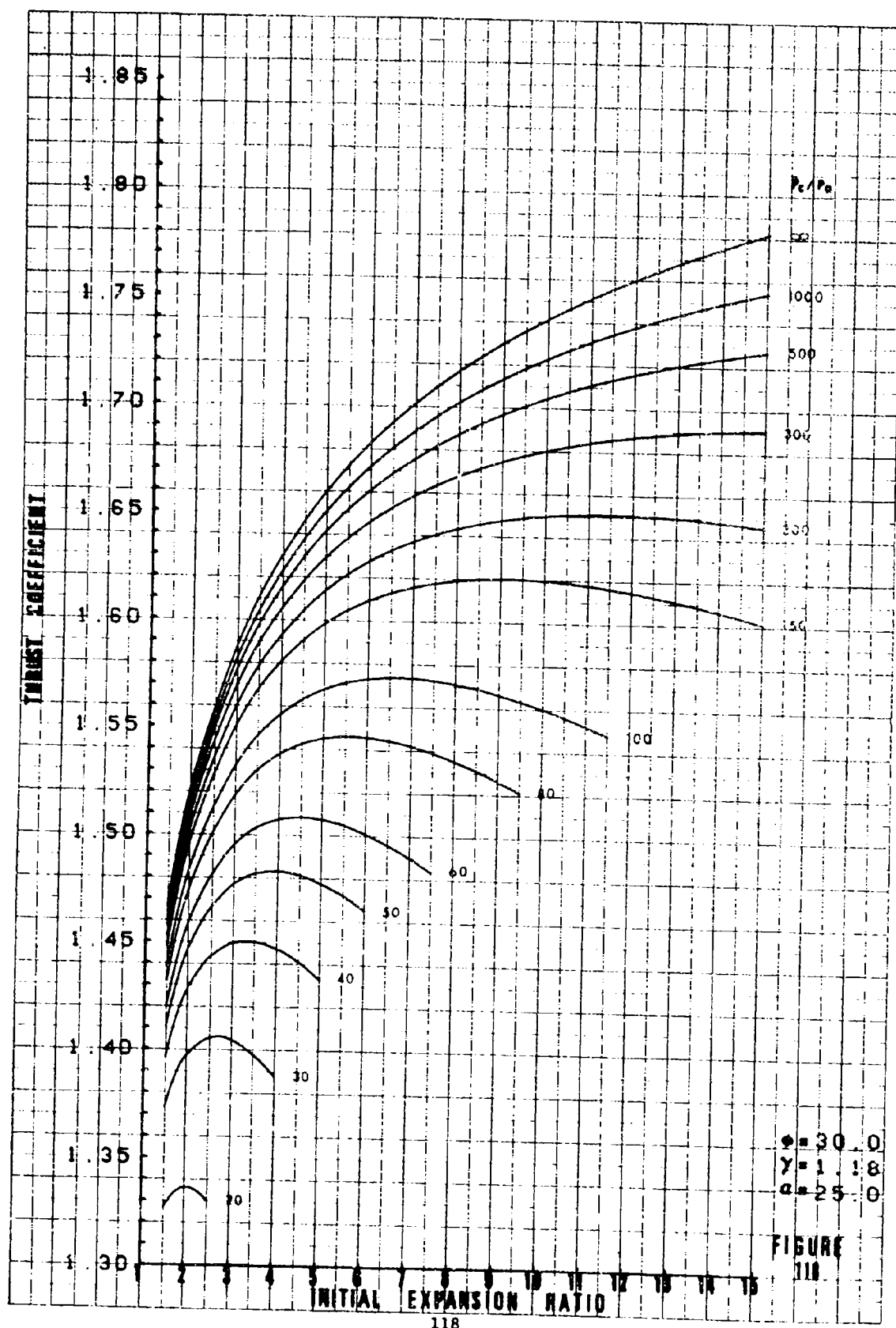


FIGURE 117



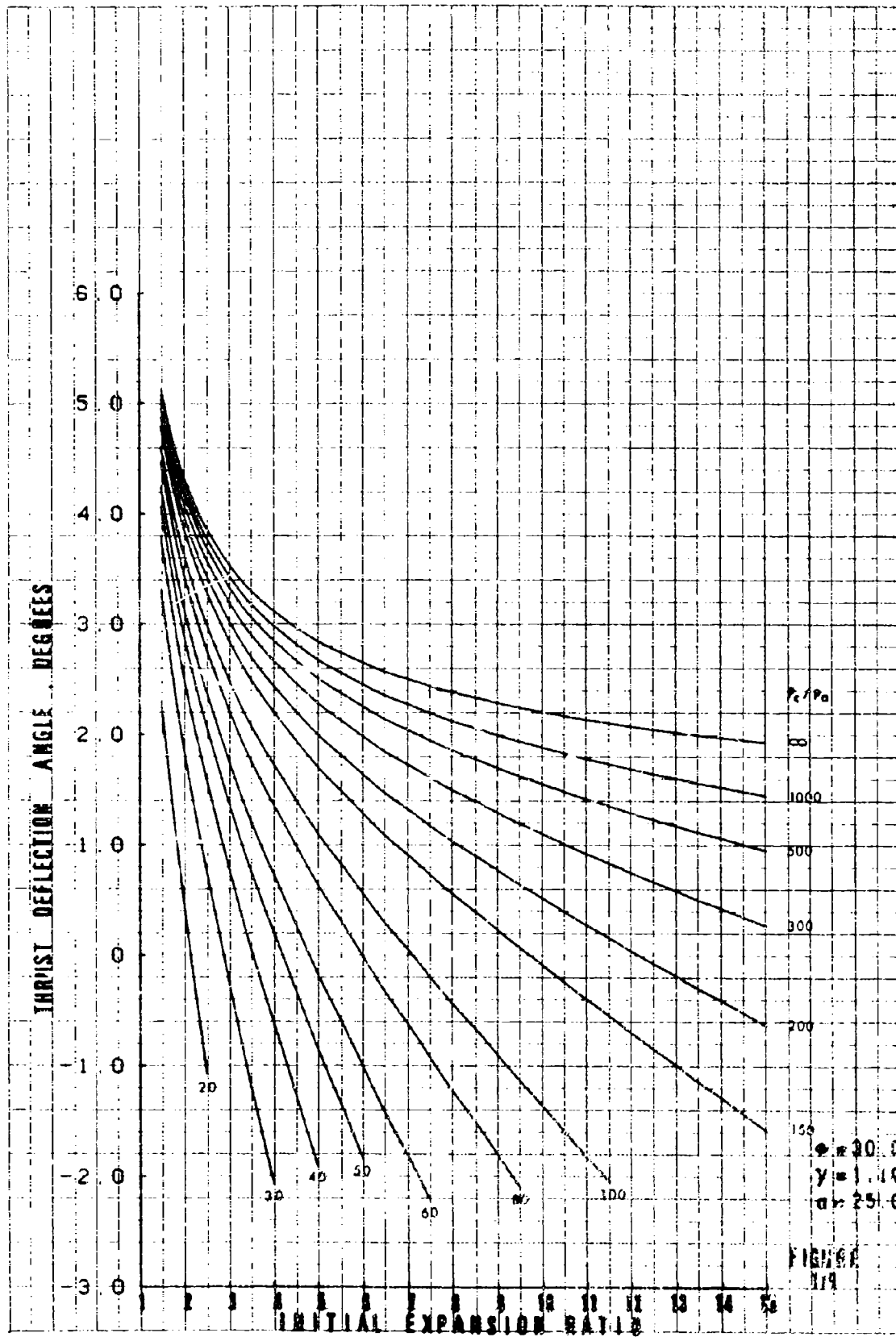
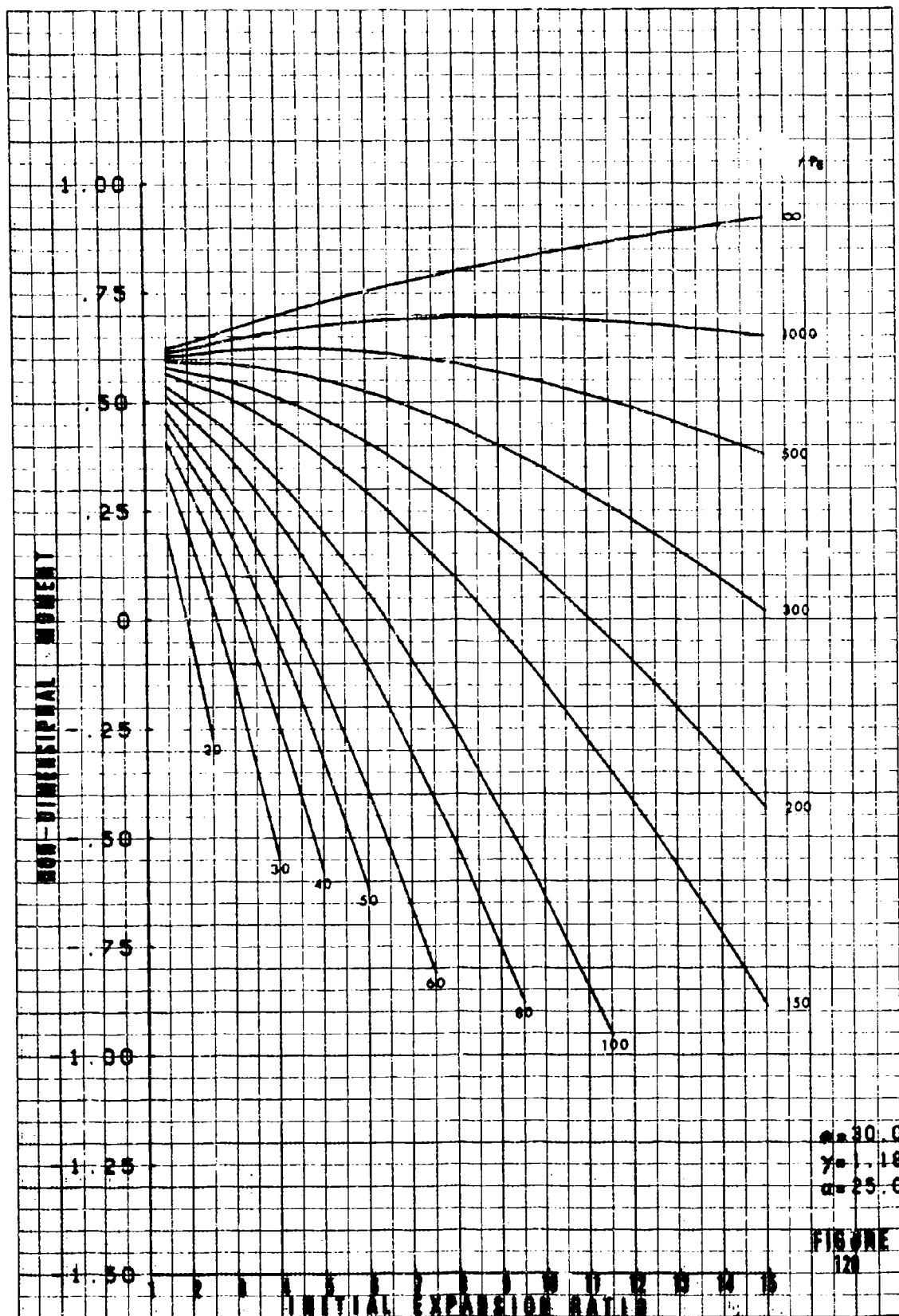
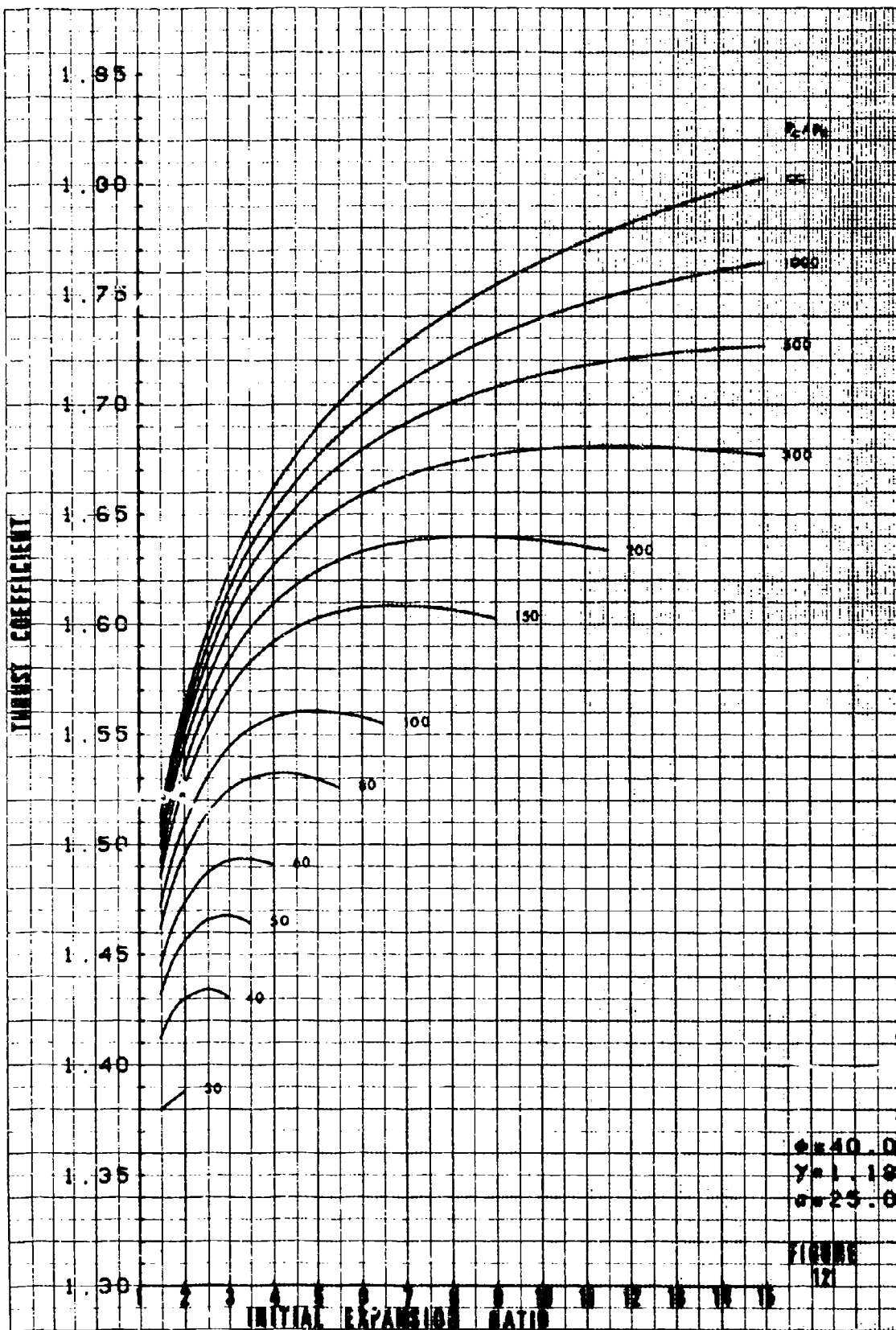
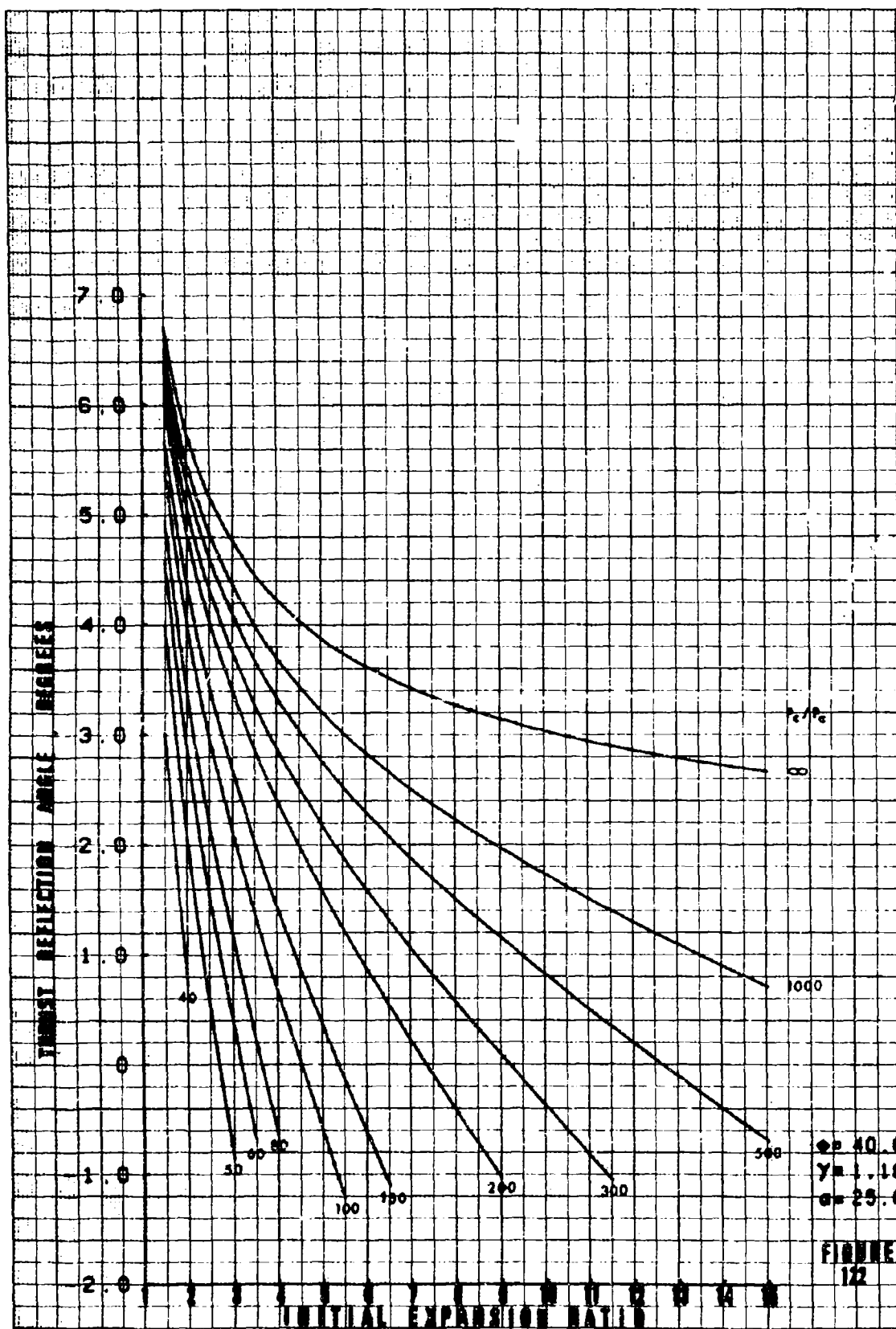
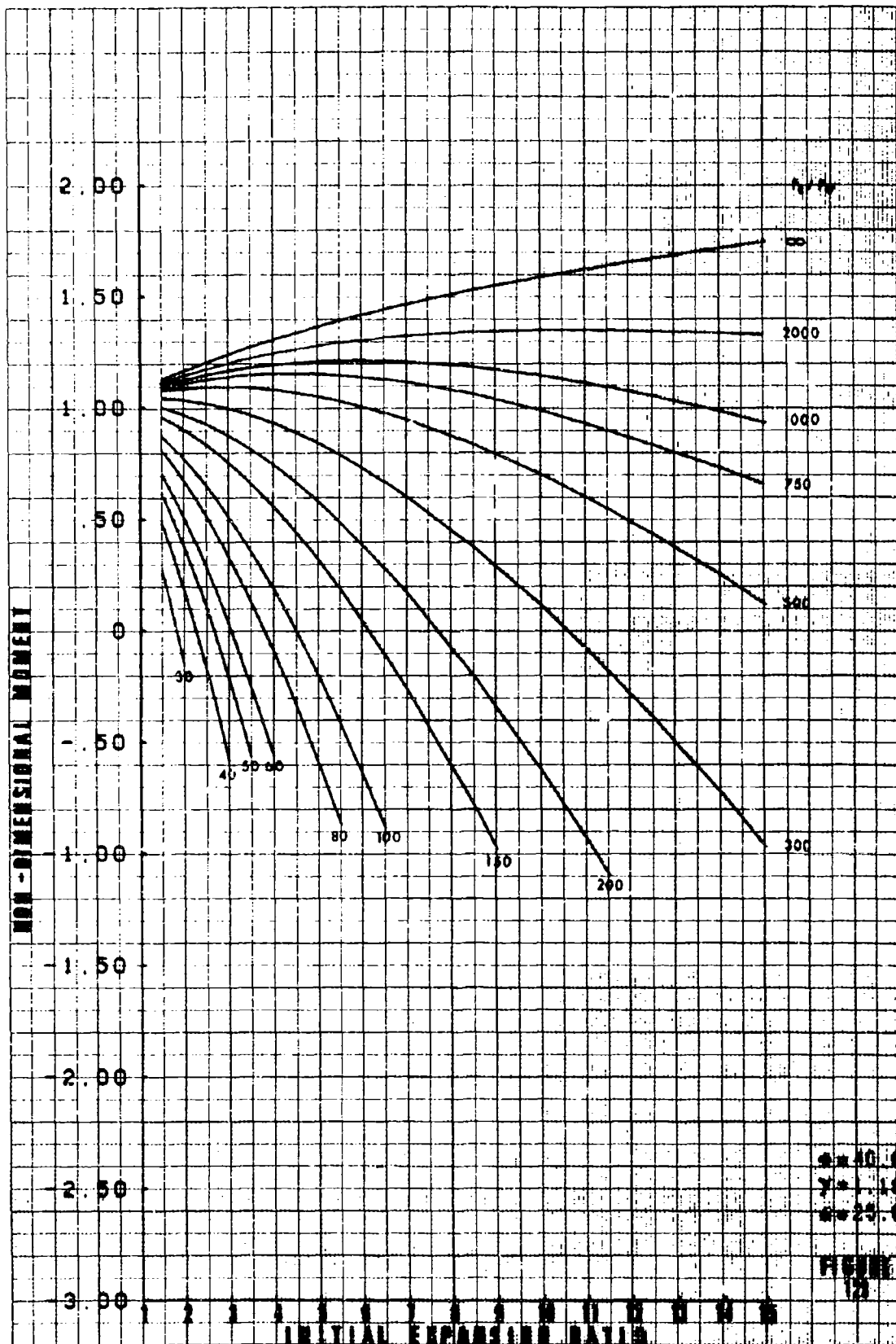


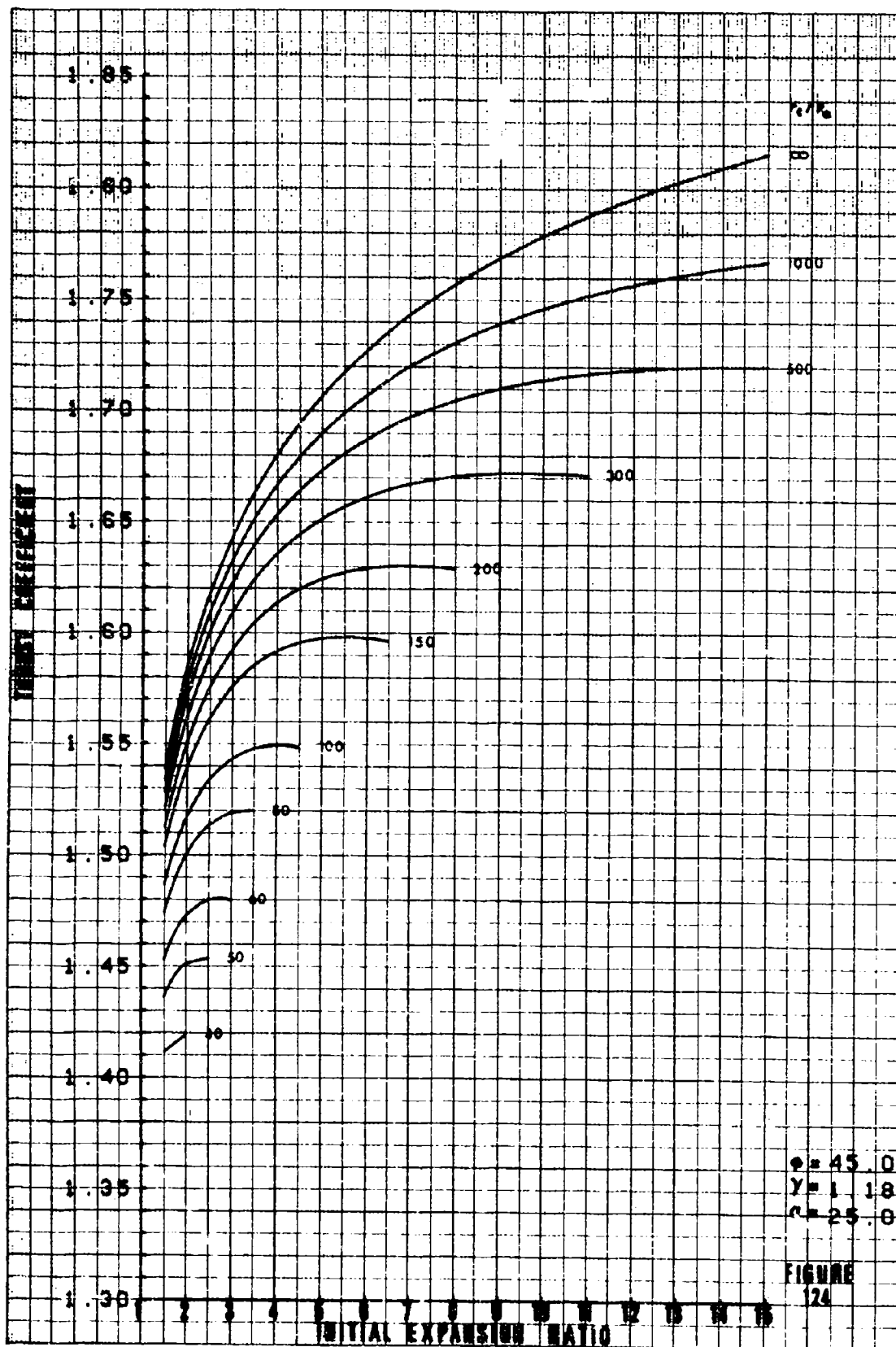
FIGURE 119











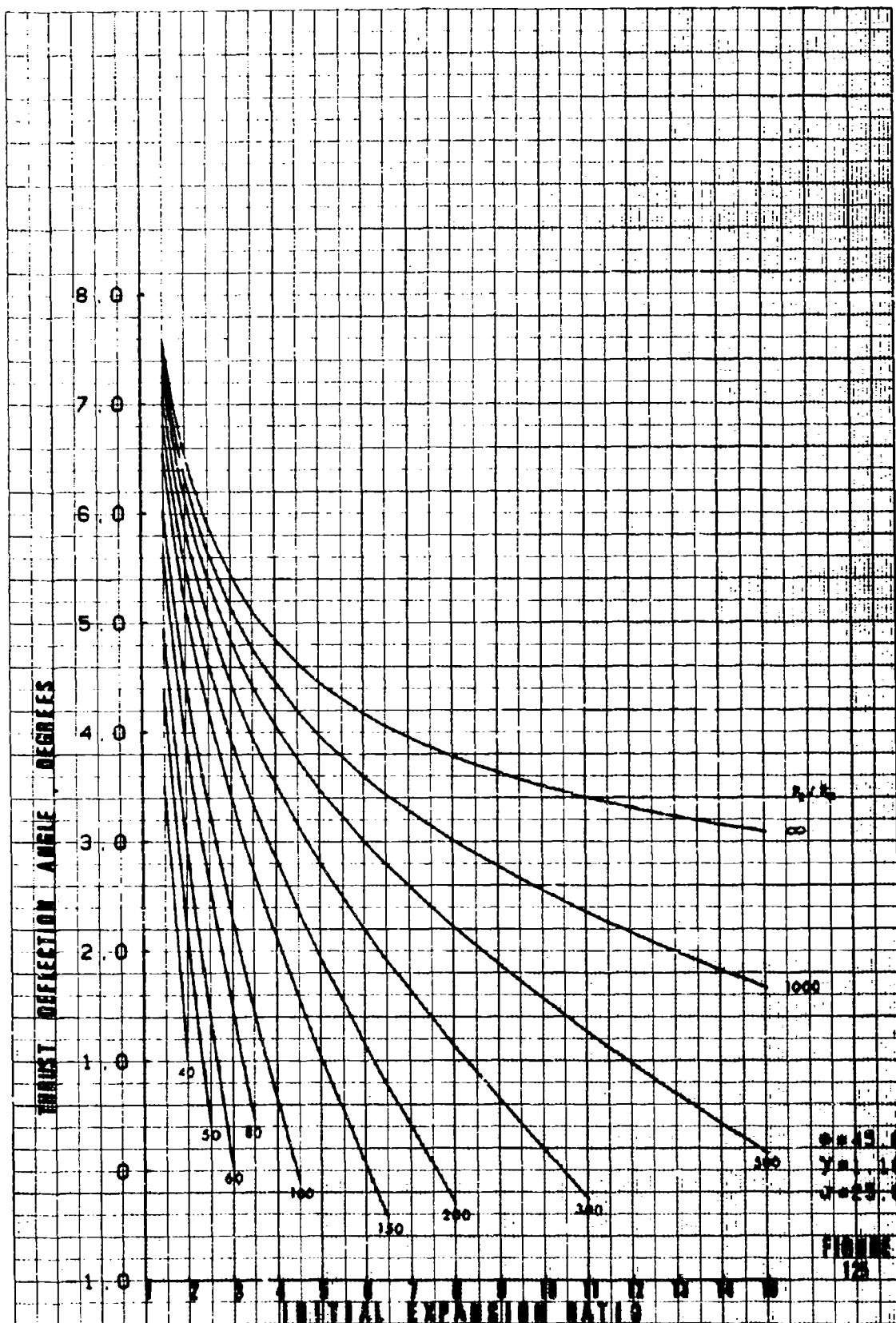
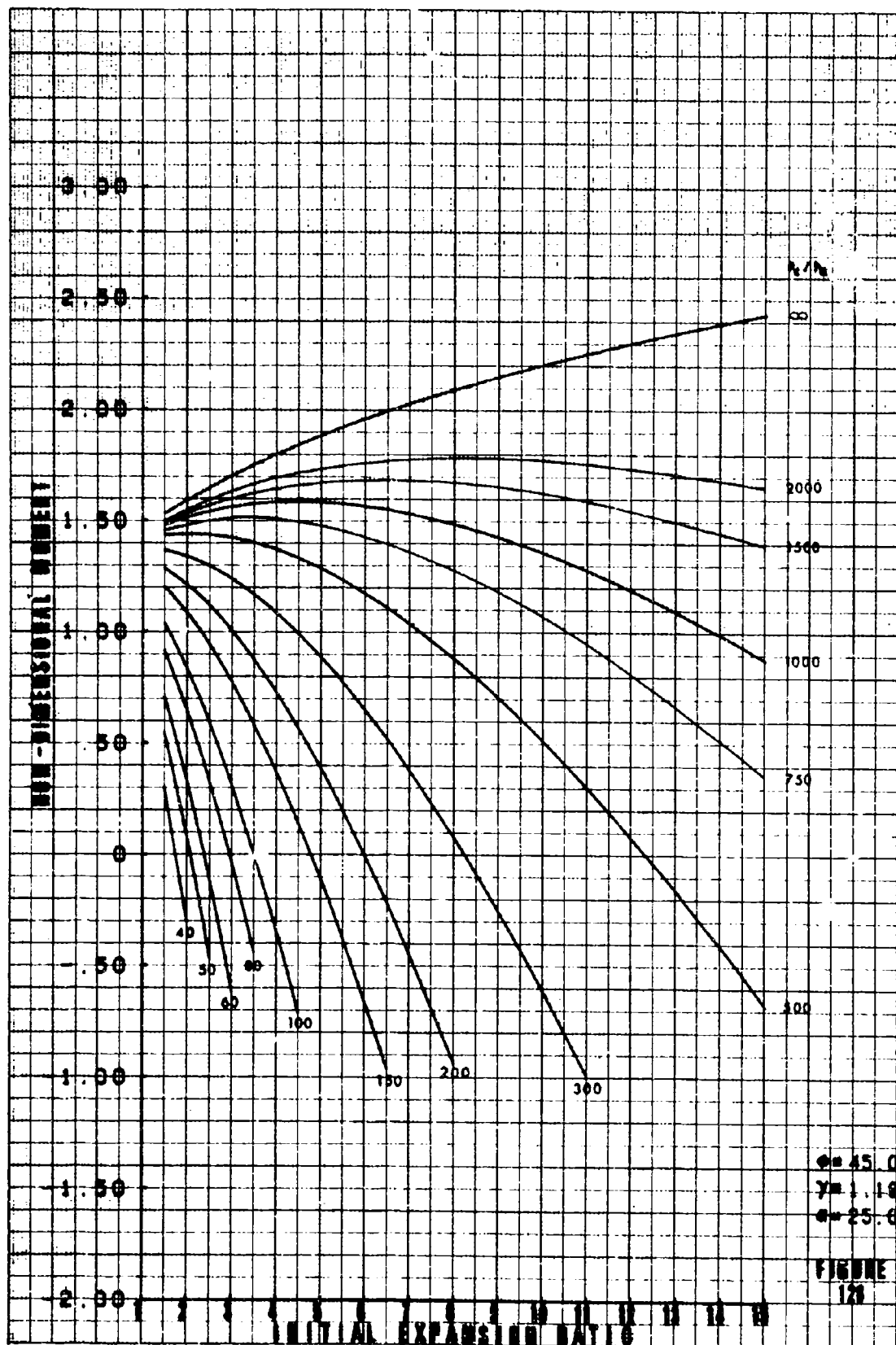
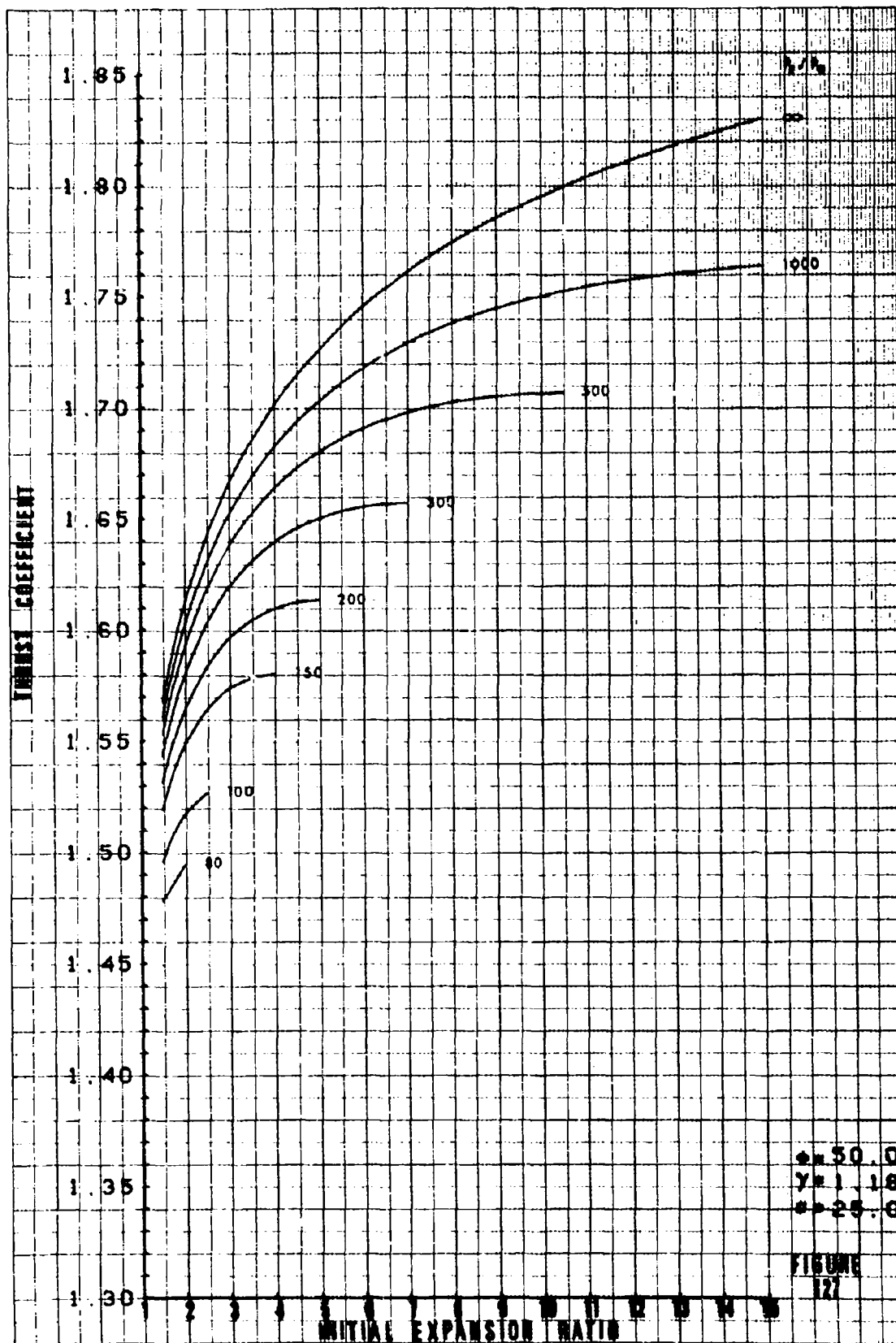


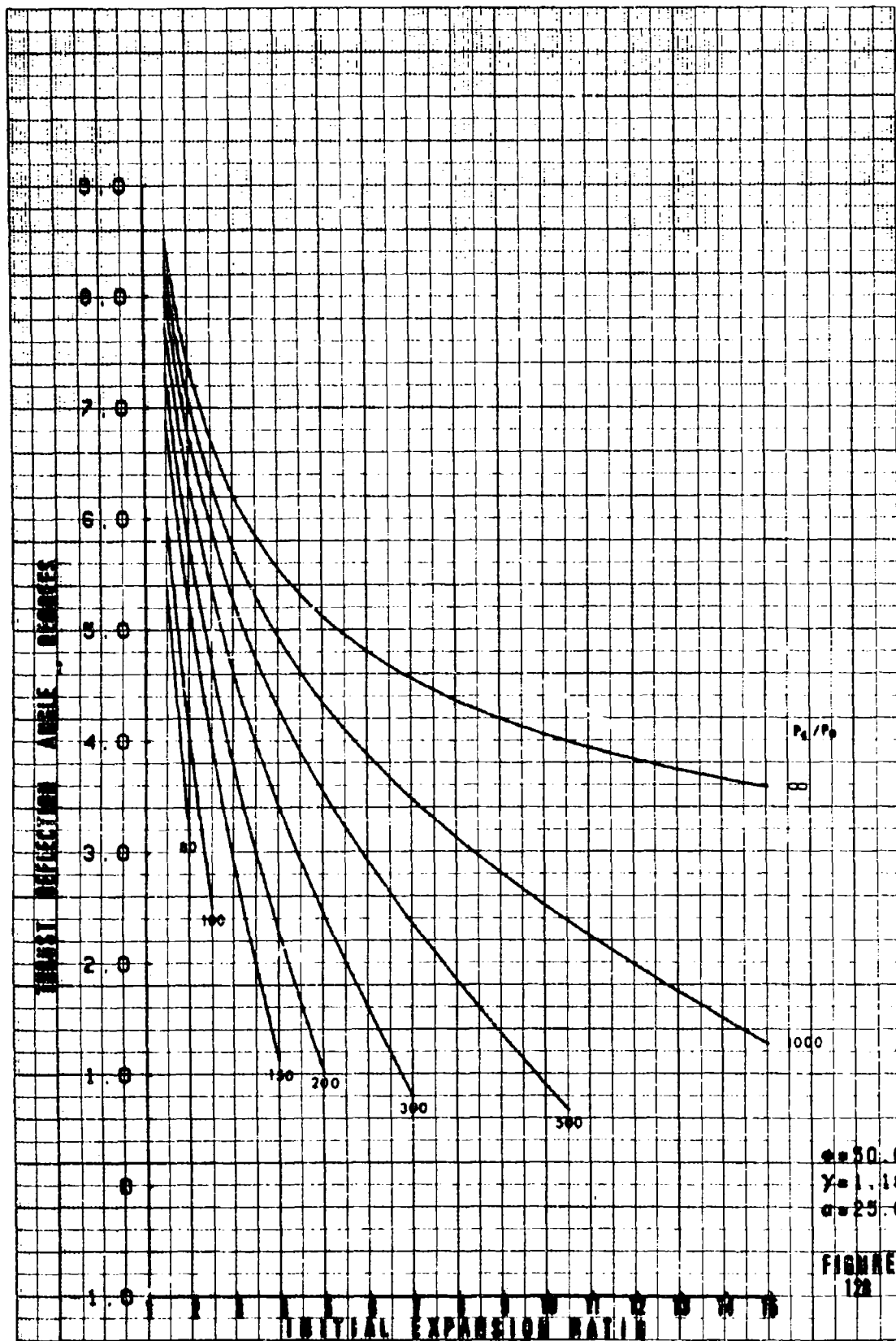
FIGURE 125

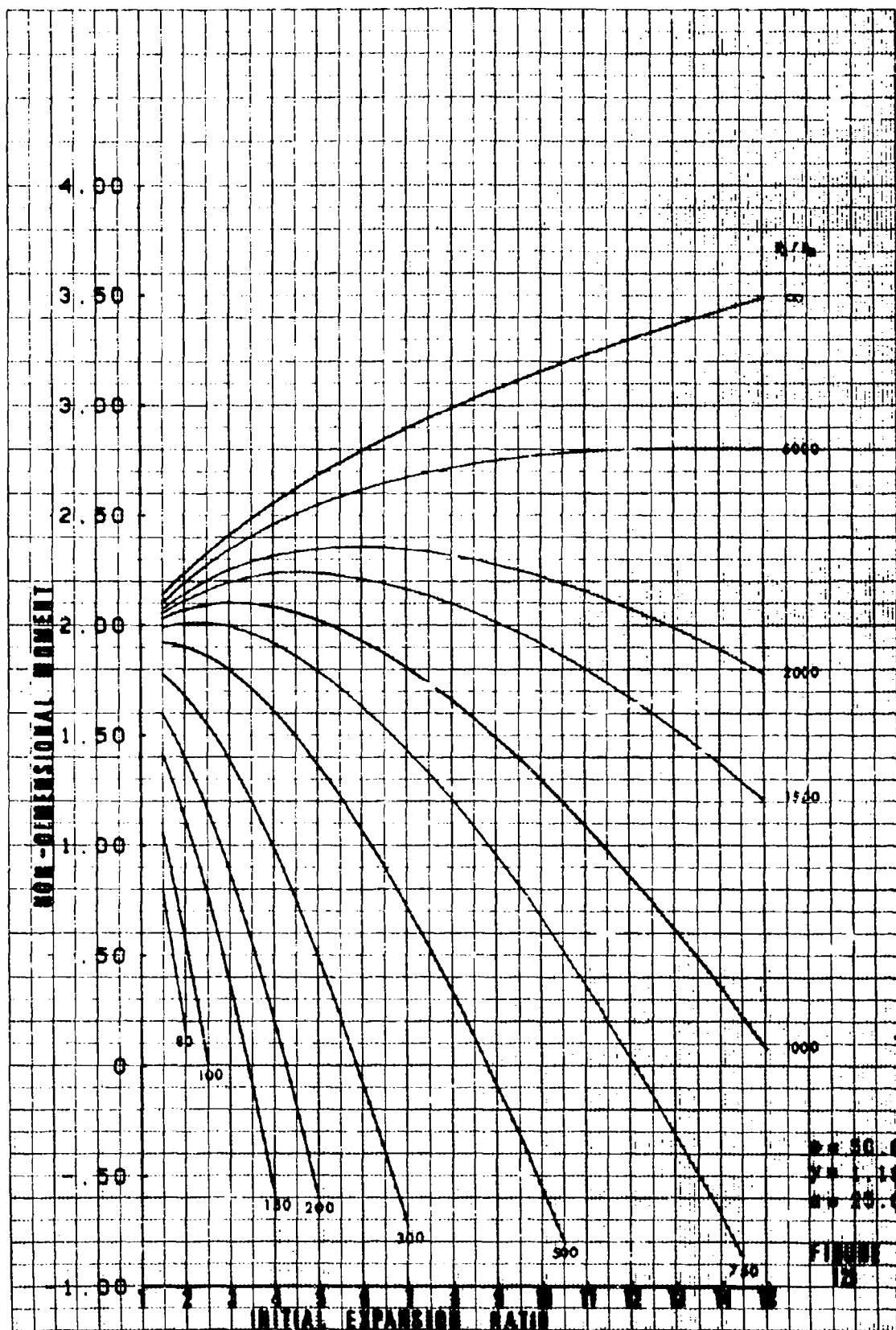


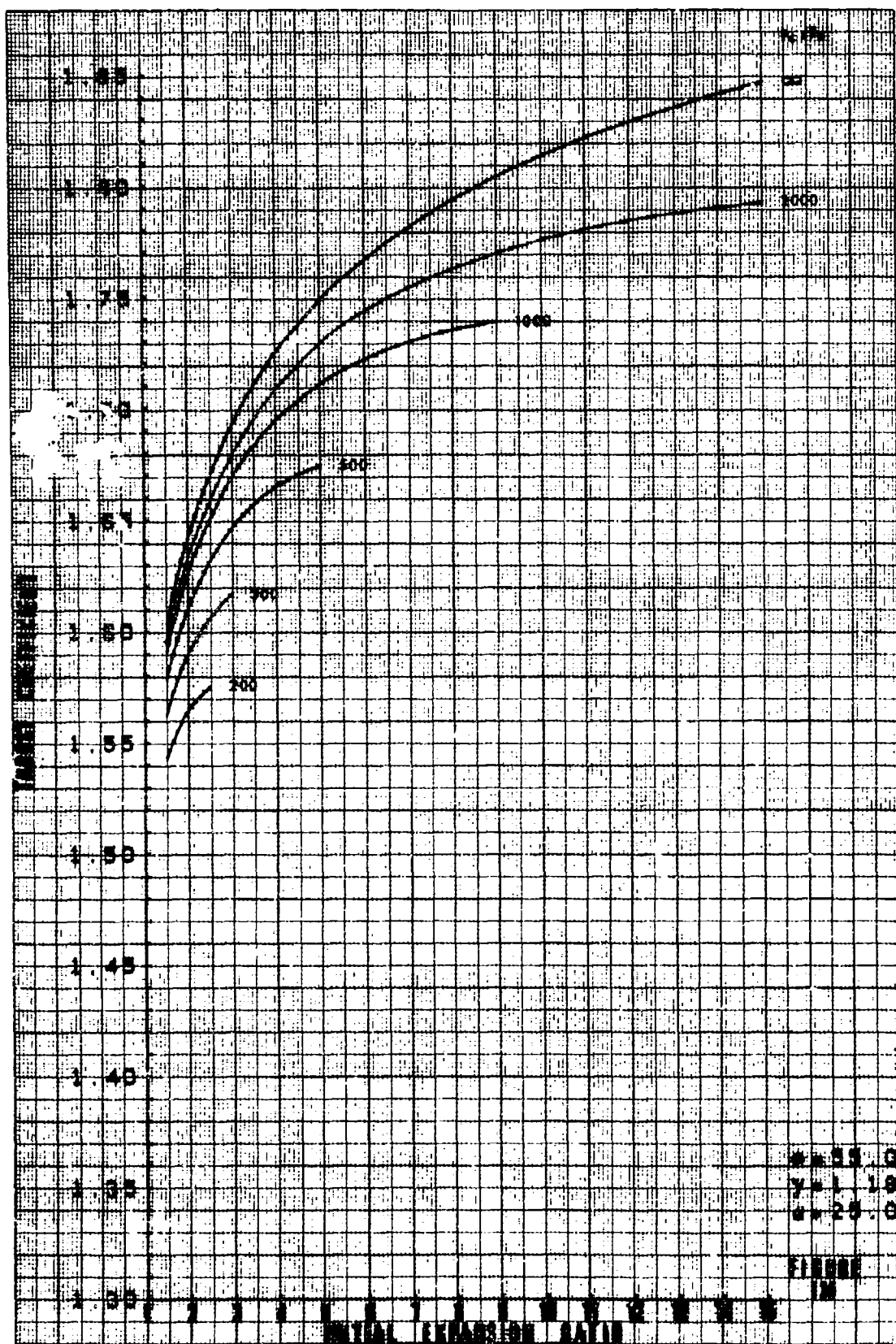


$\gamma = 50.0$
 $\gamma = 1.18$
 $\gamma = 25.0$

FIGURE 127







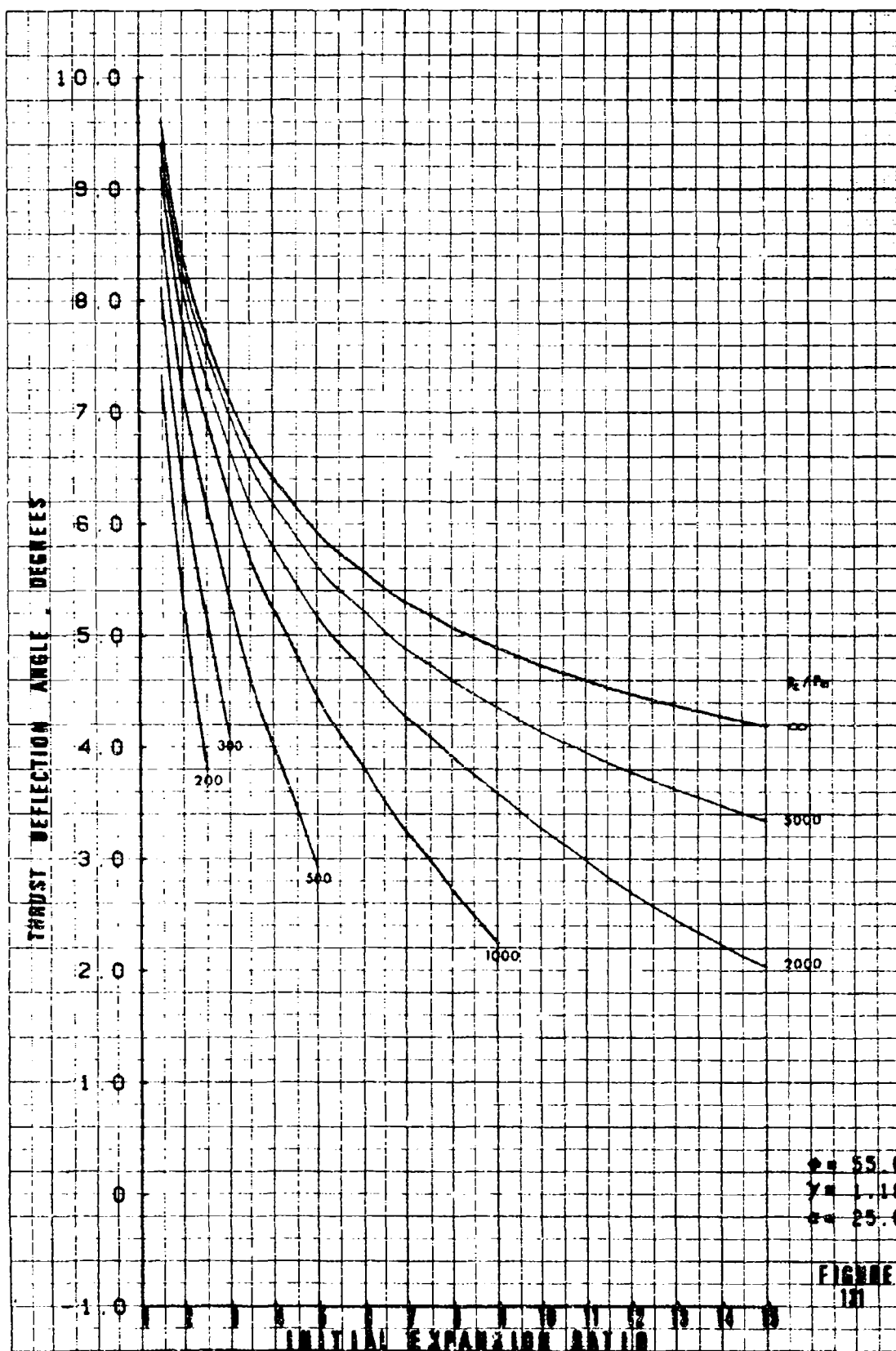
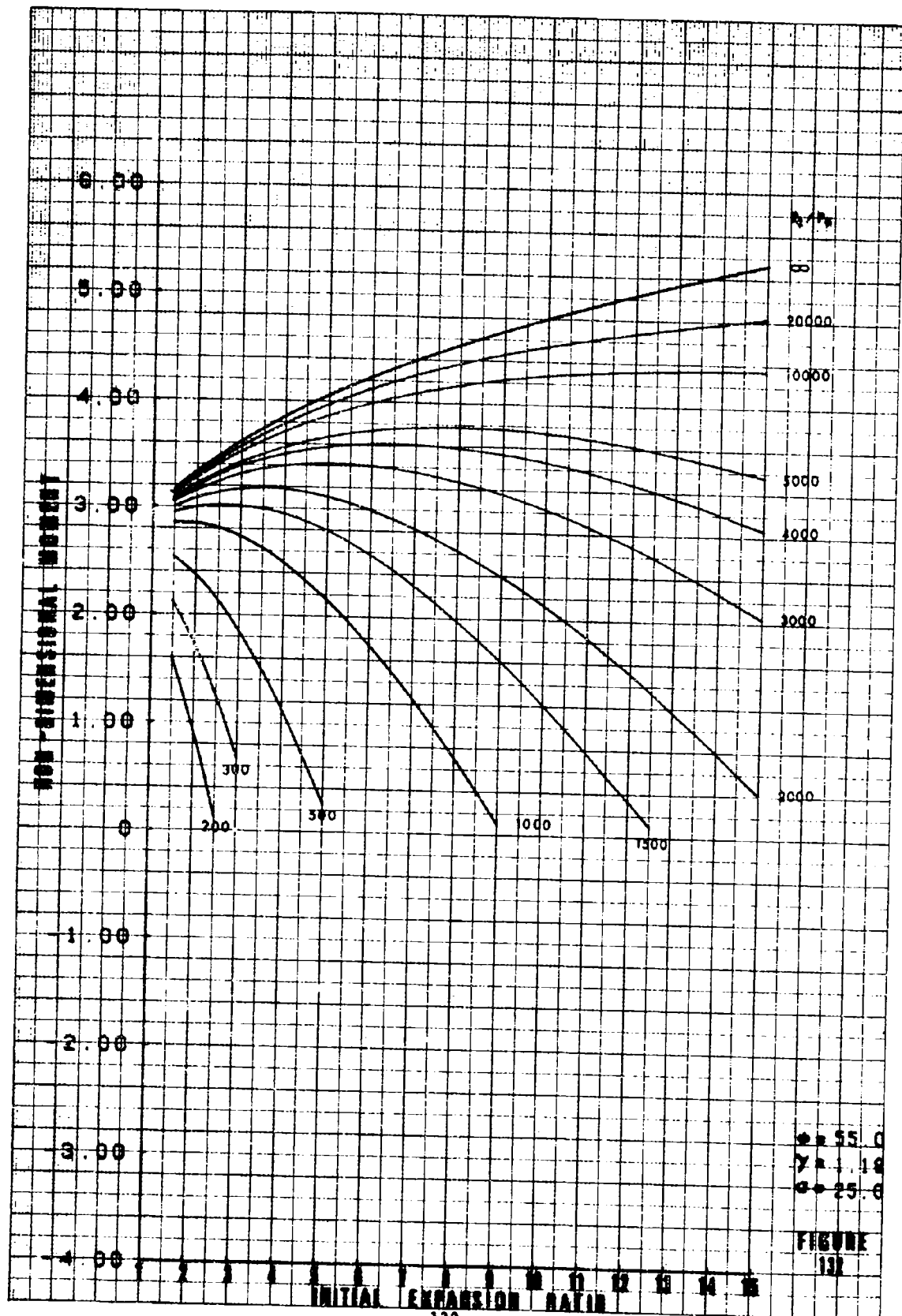
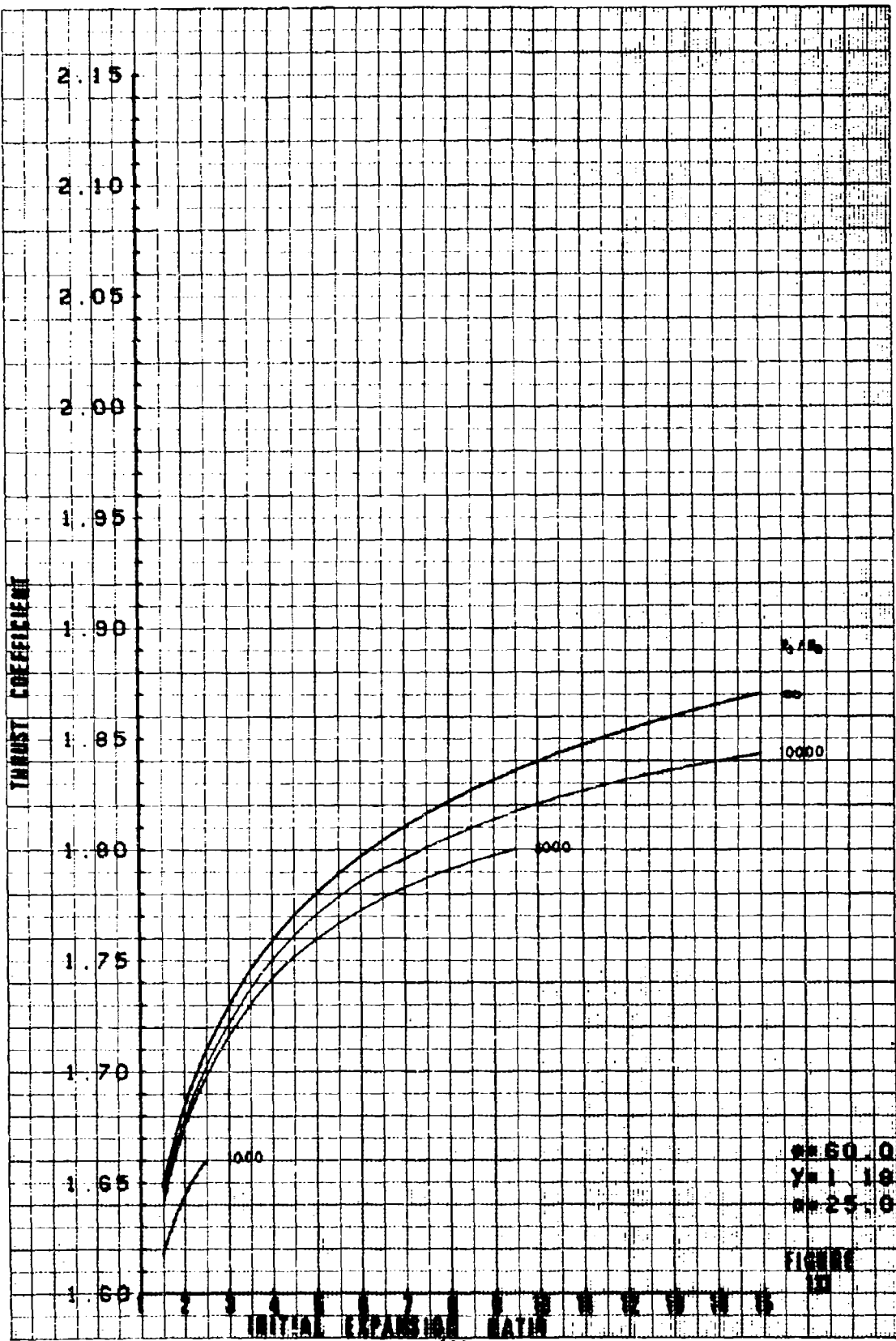


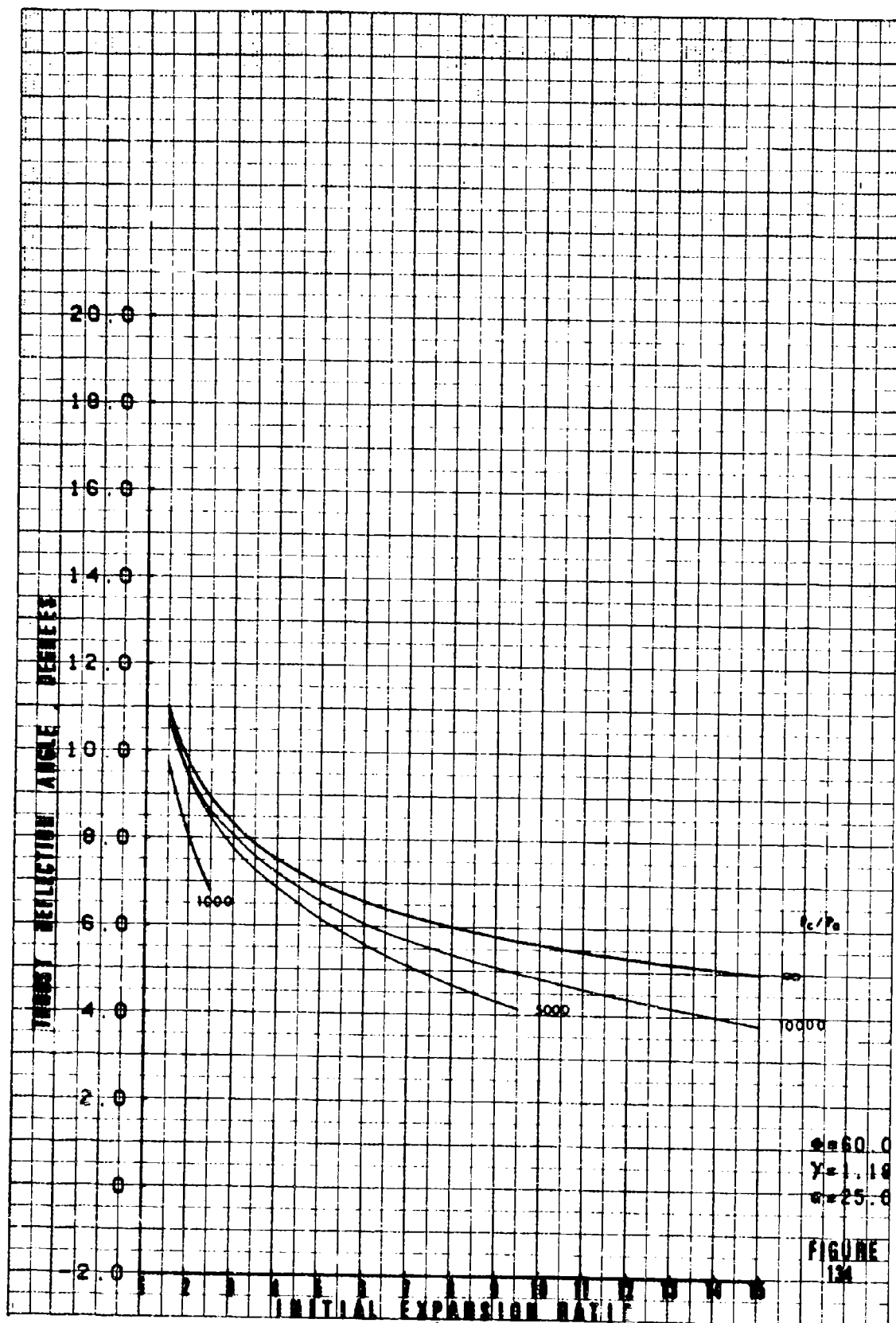
FIGURE 121

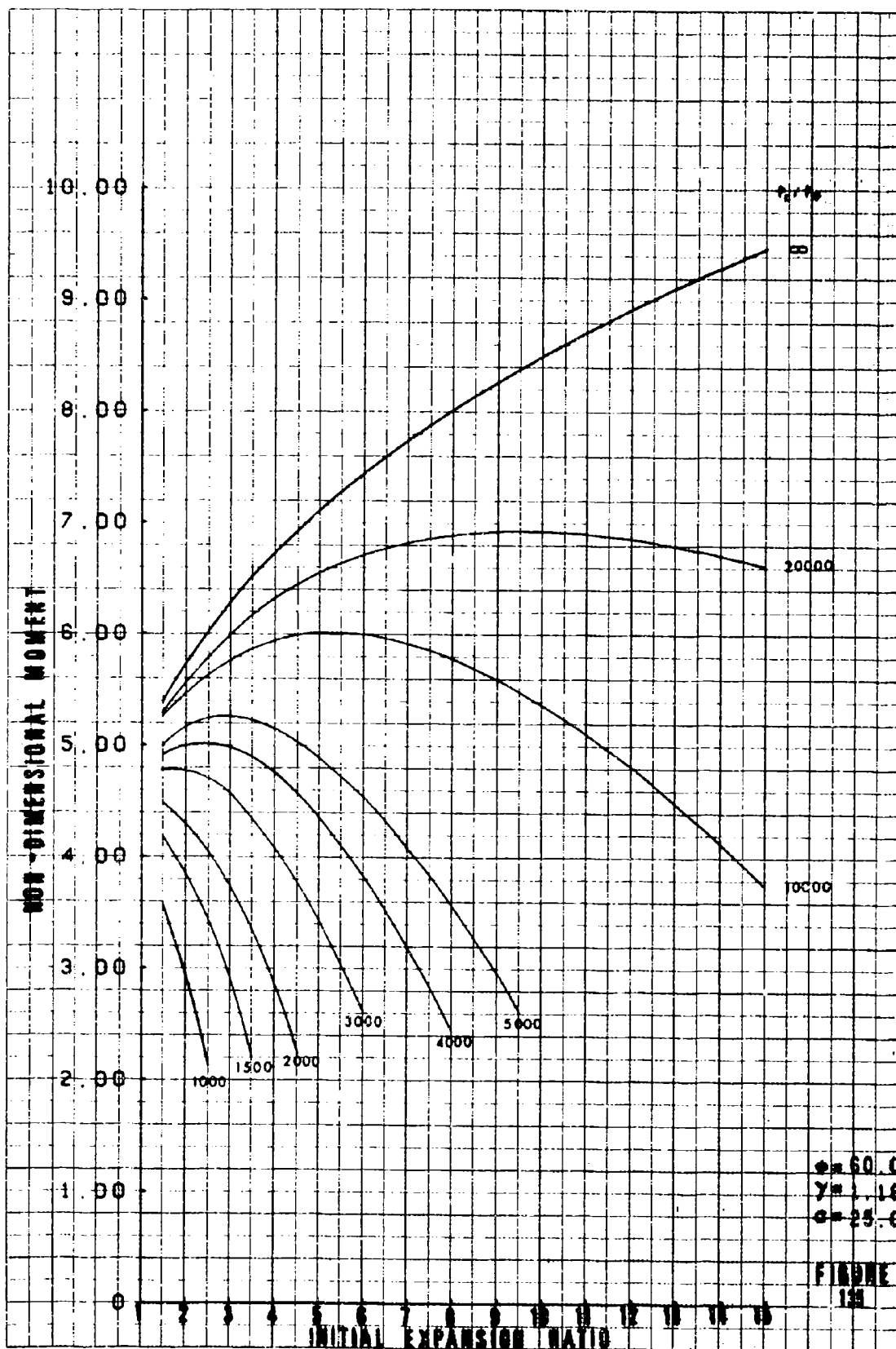


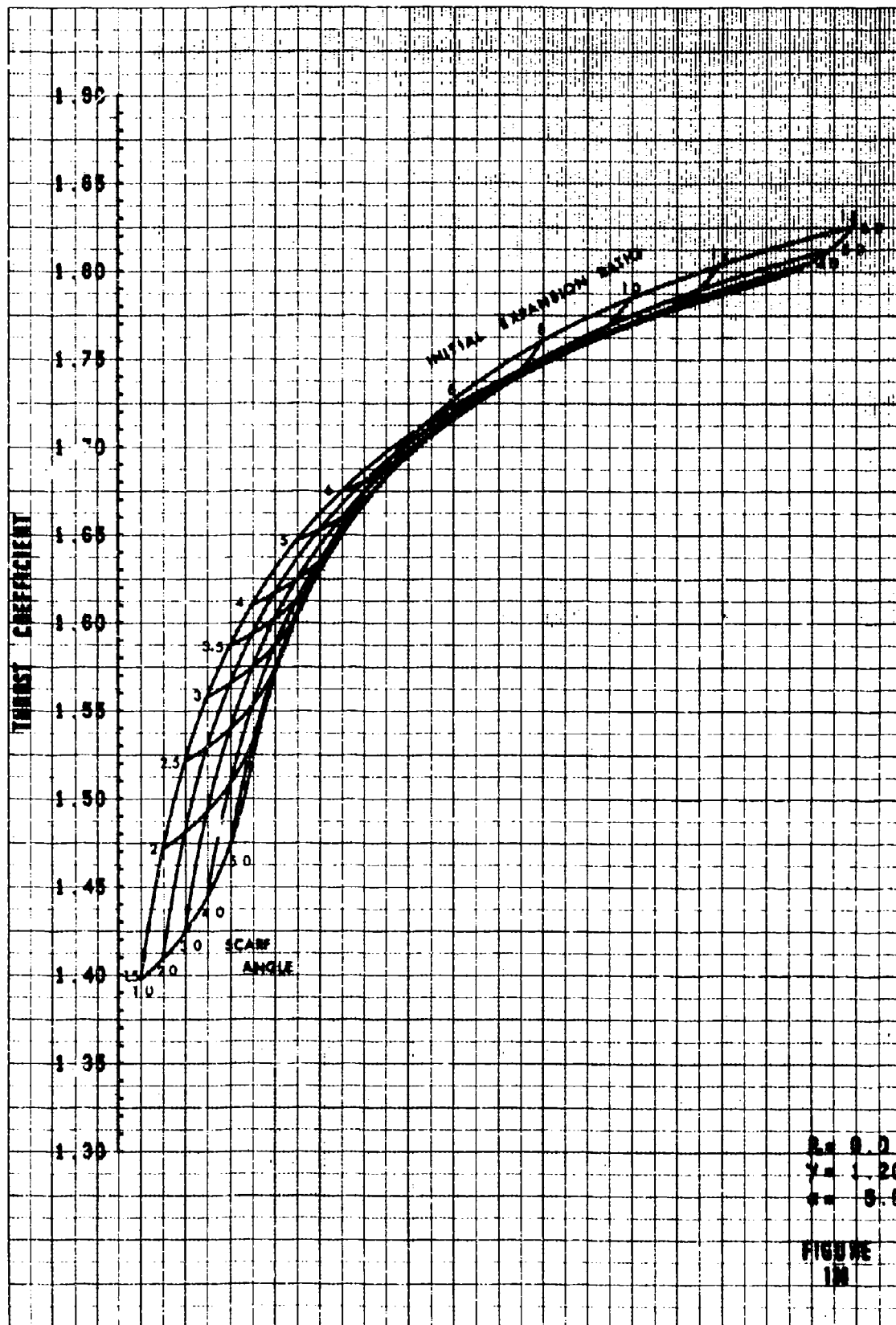


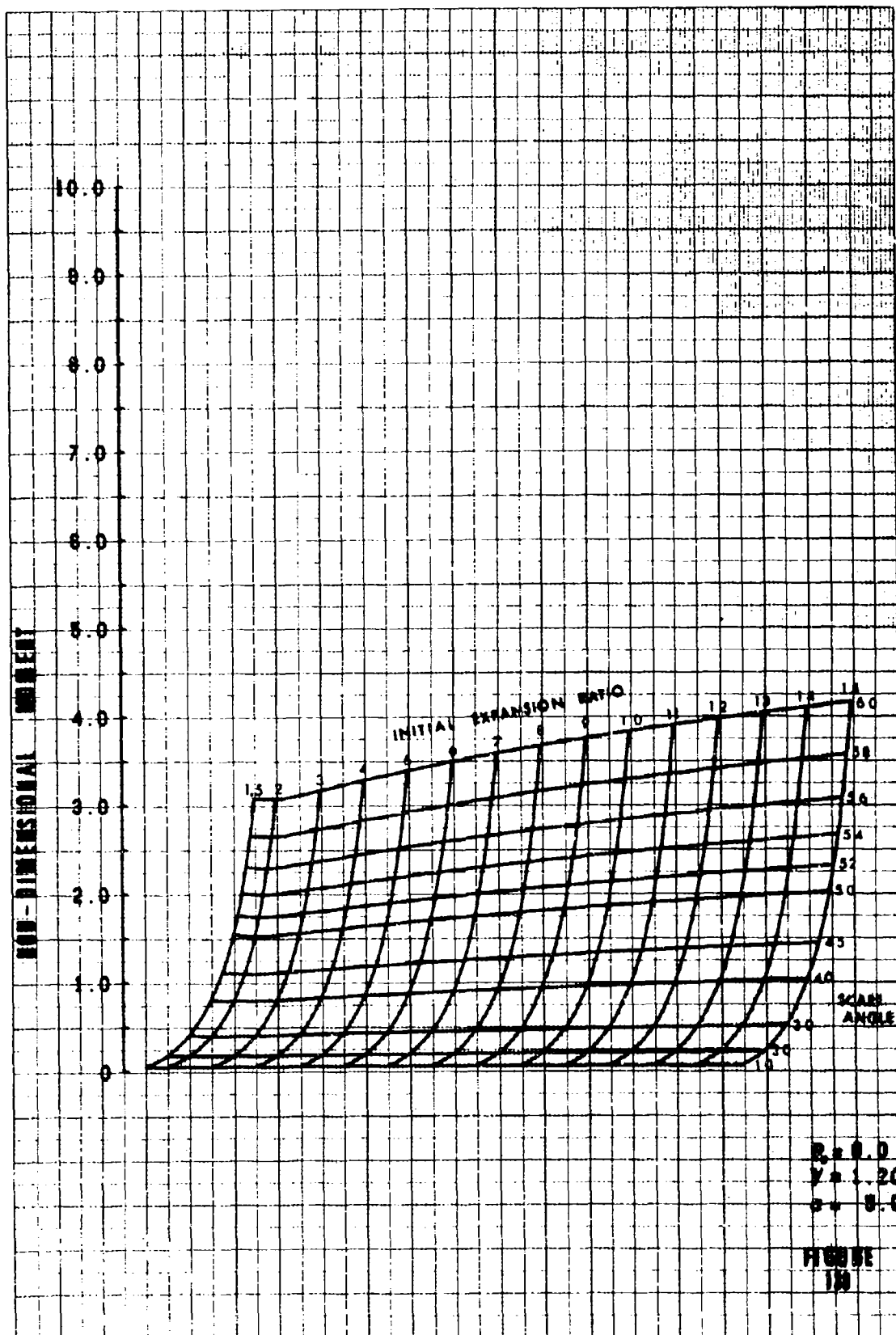
$M=60.0$
 $M=11.8$
 $M=25.0$

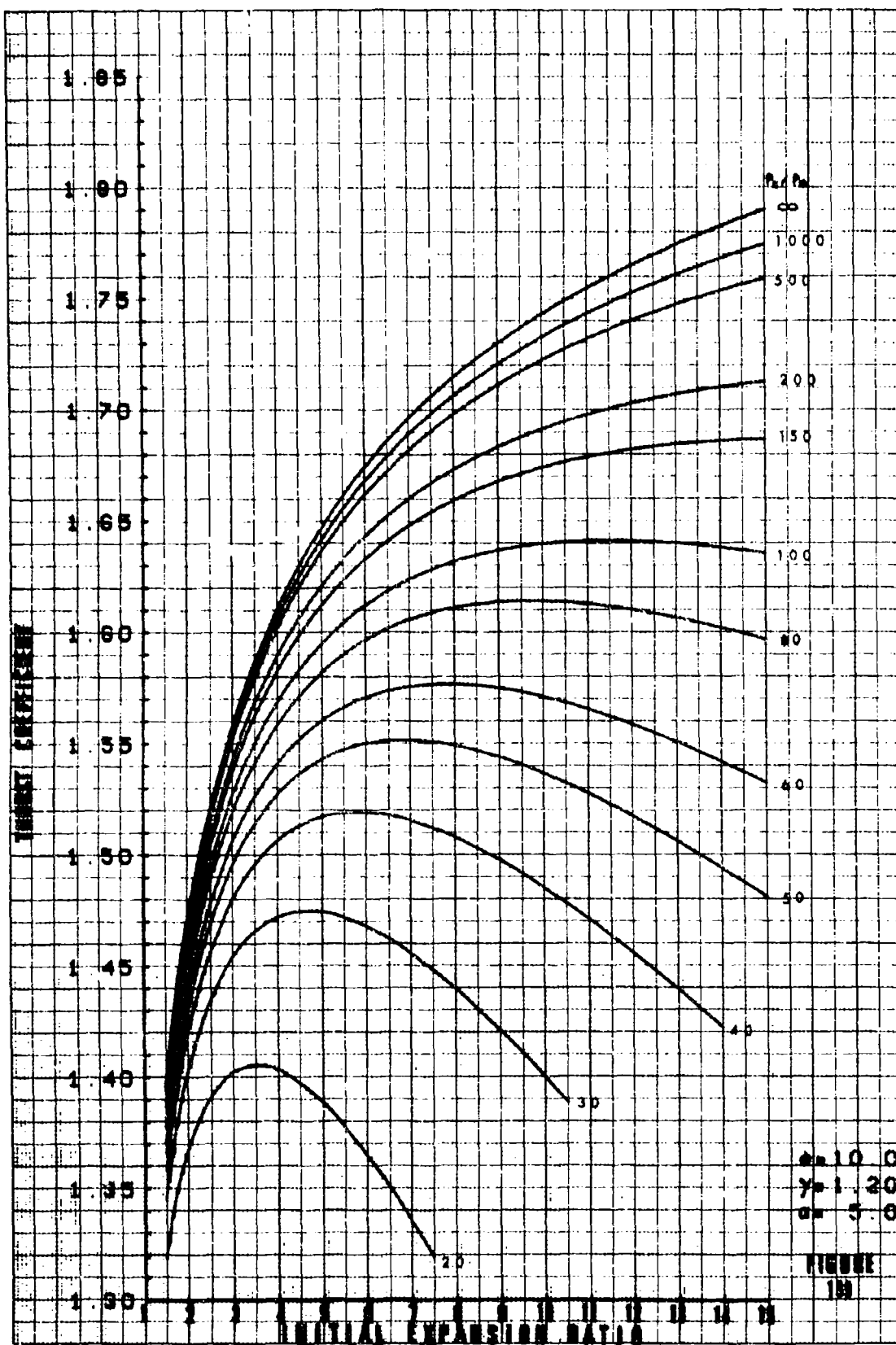
FIGURE 12





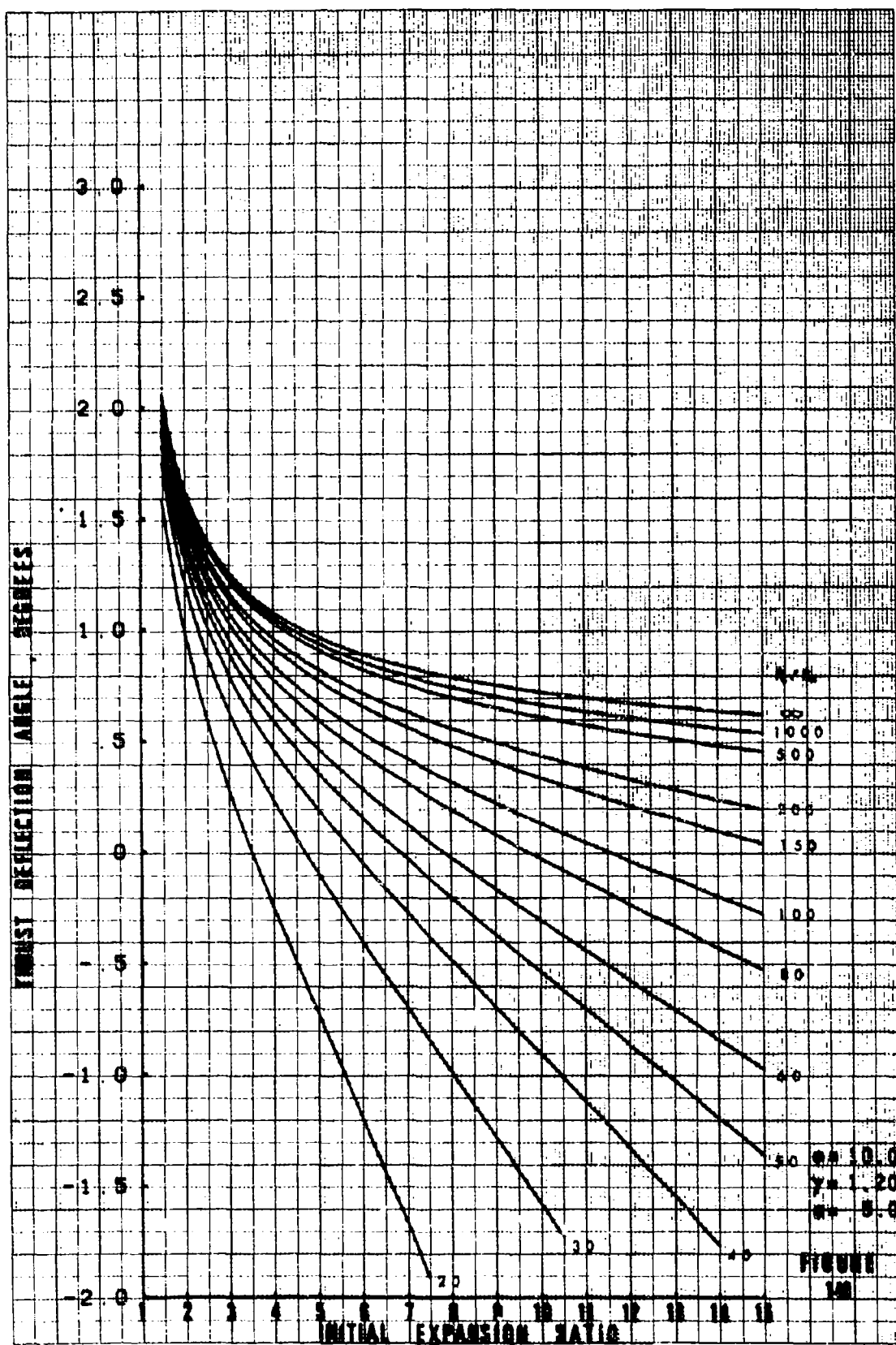






$\gamma = 1.20$
 $g = 3.0$

FIGURE 100



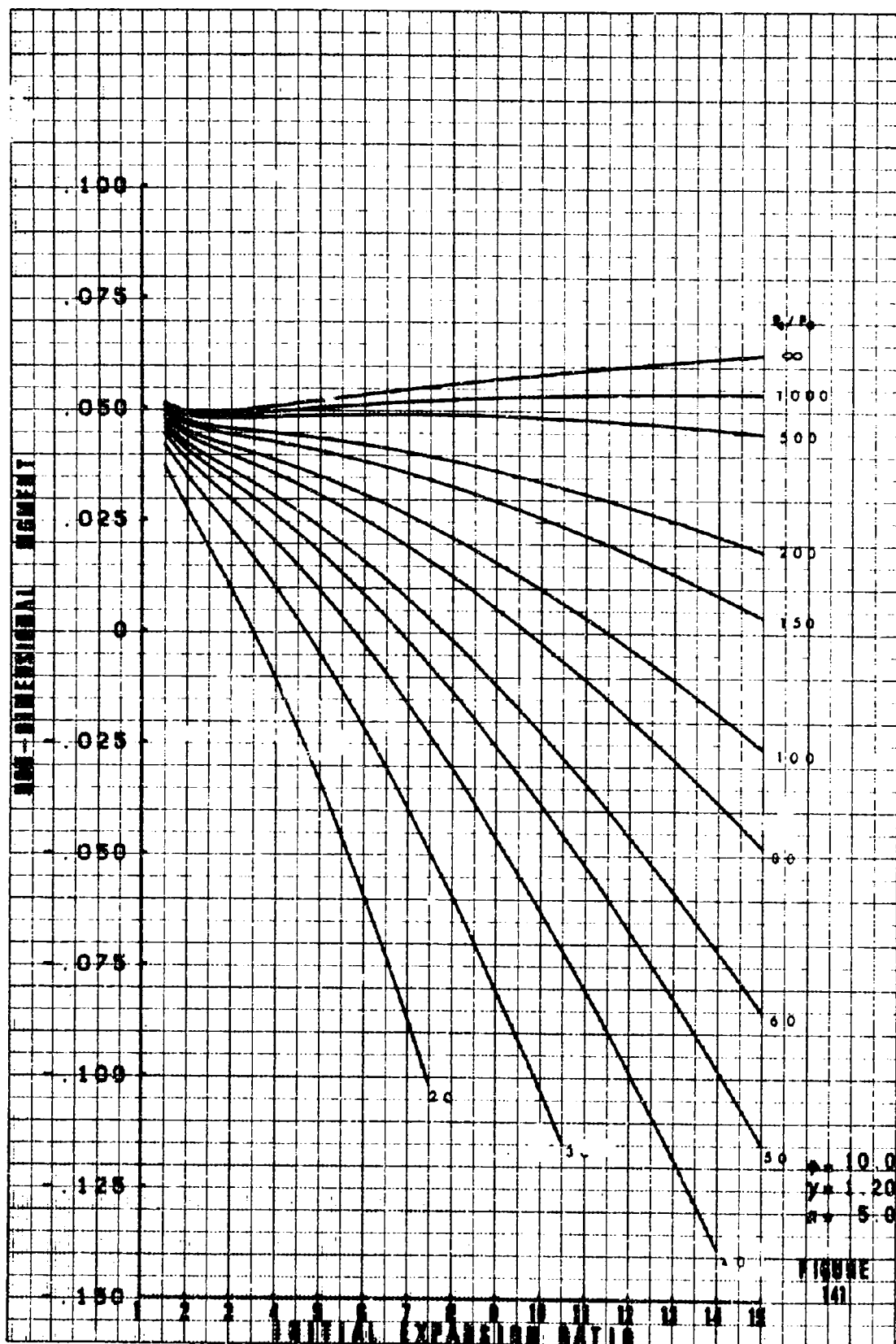
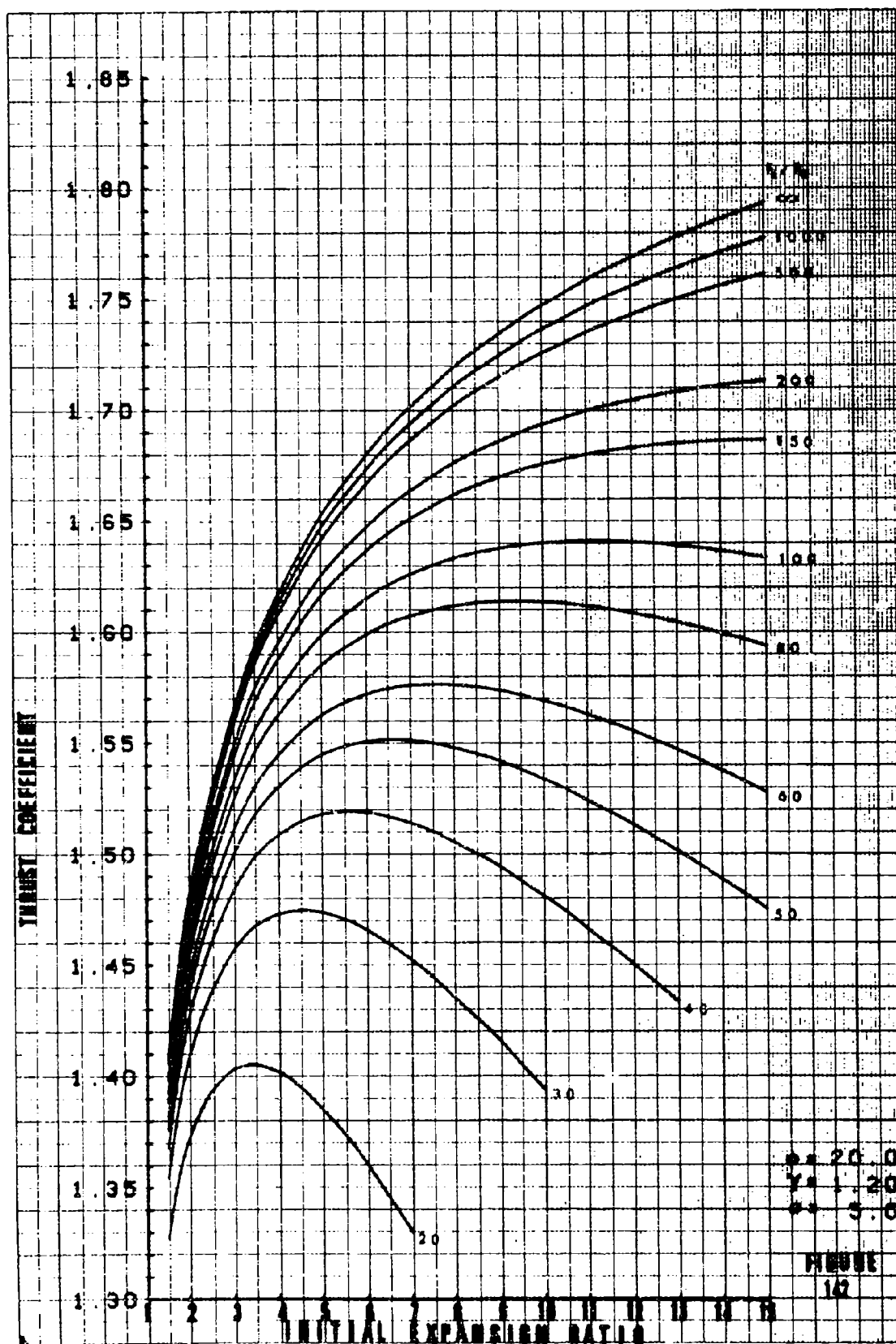
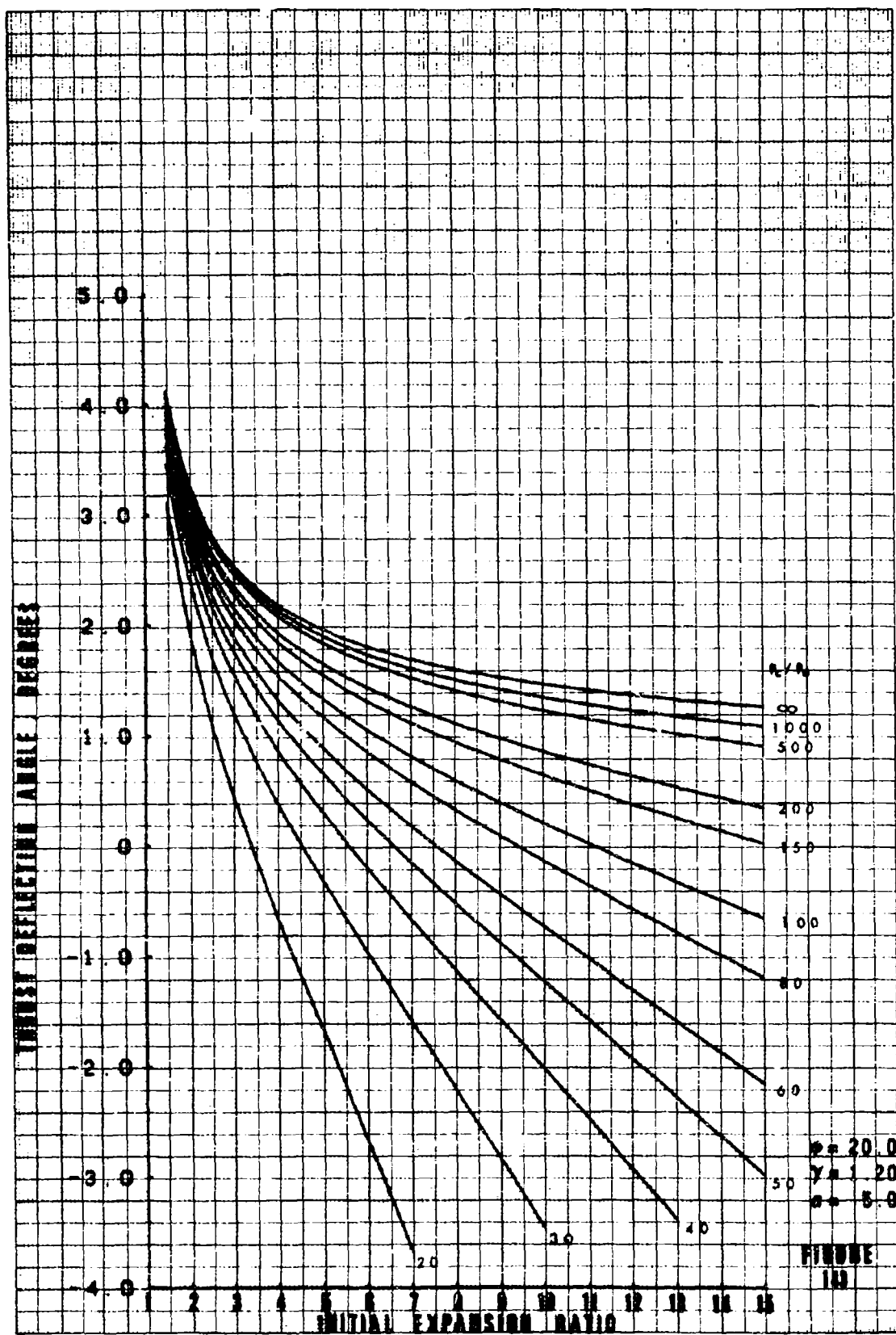
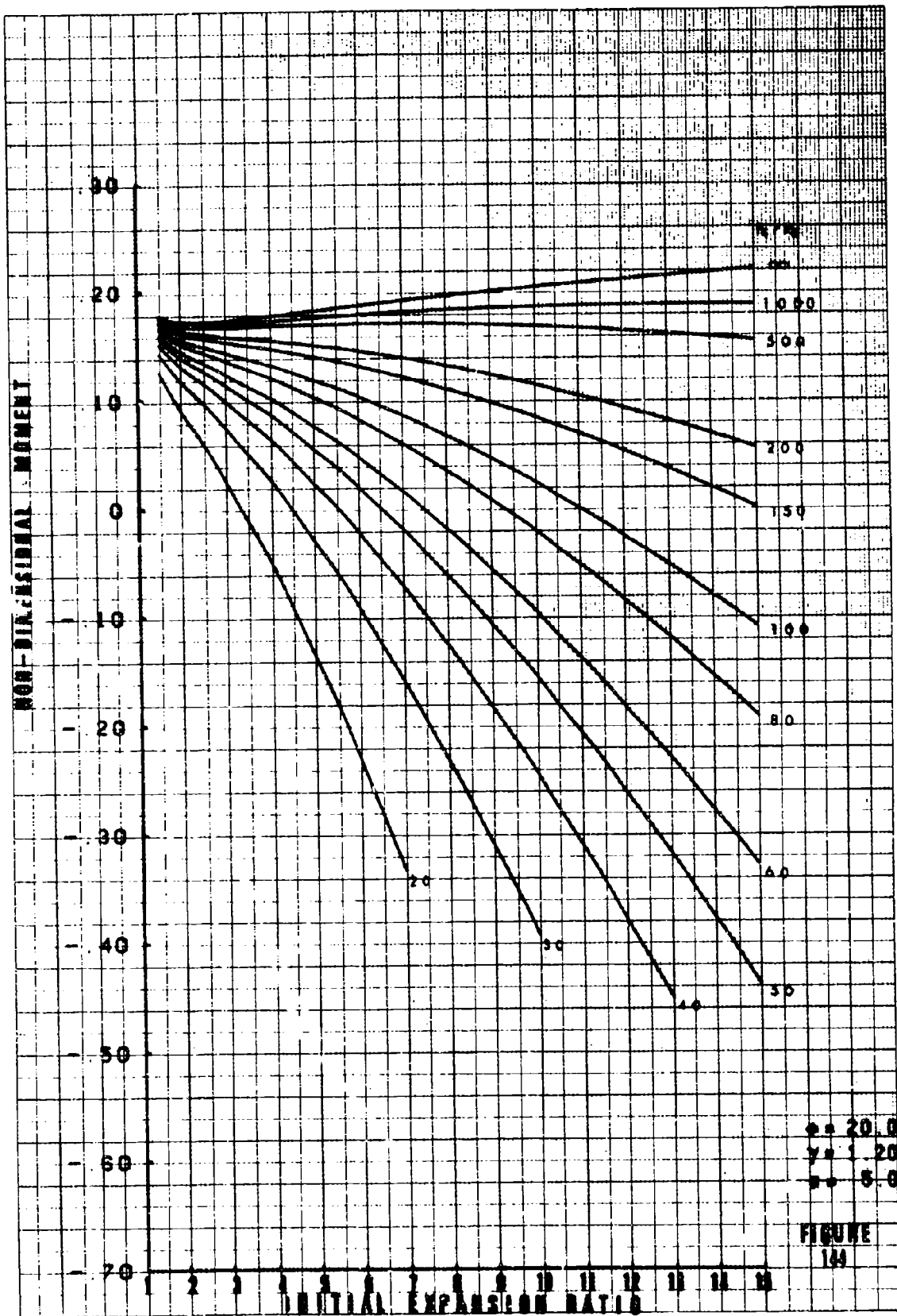
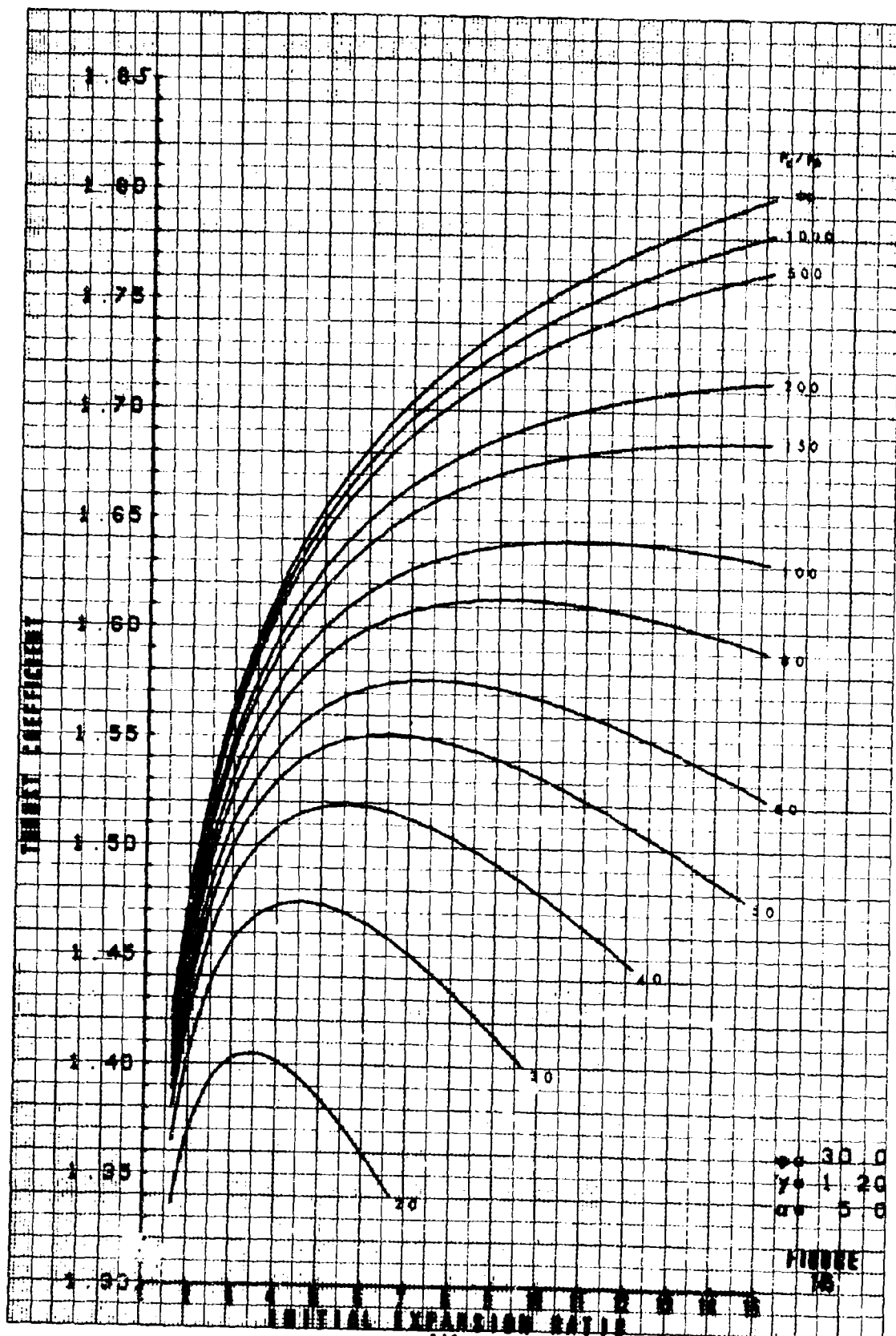


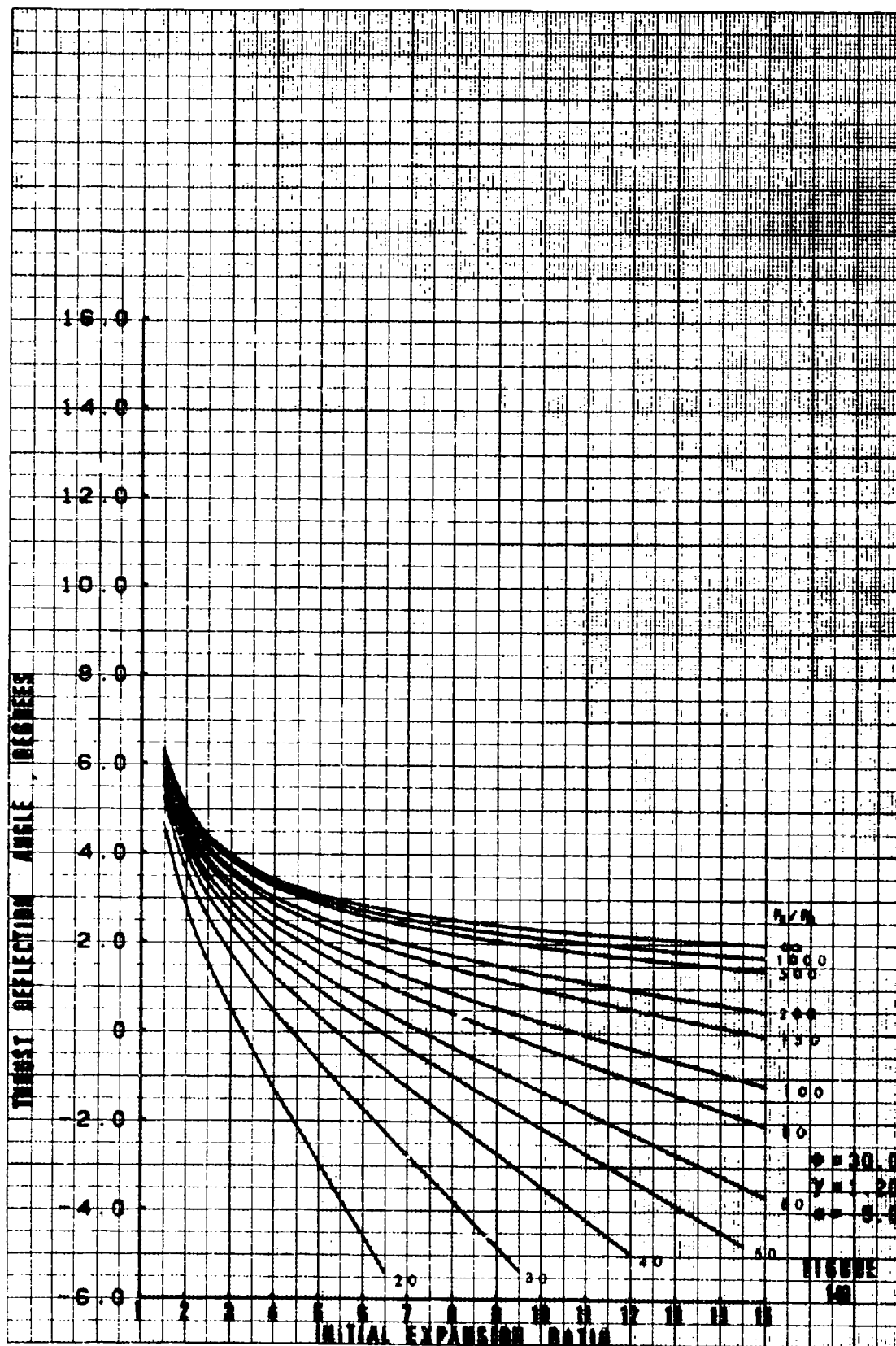
FIGURE 141

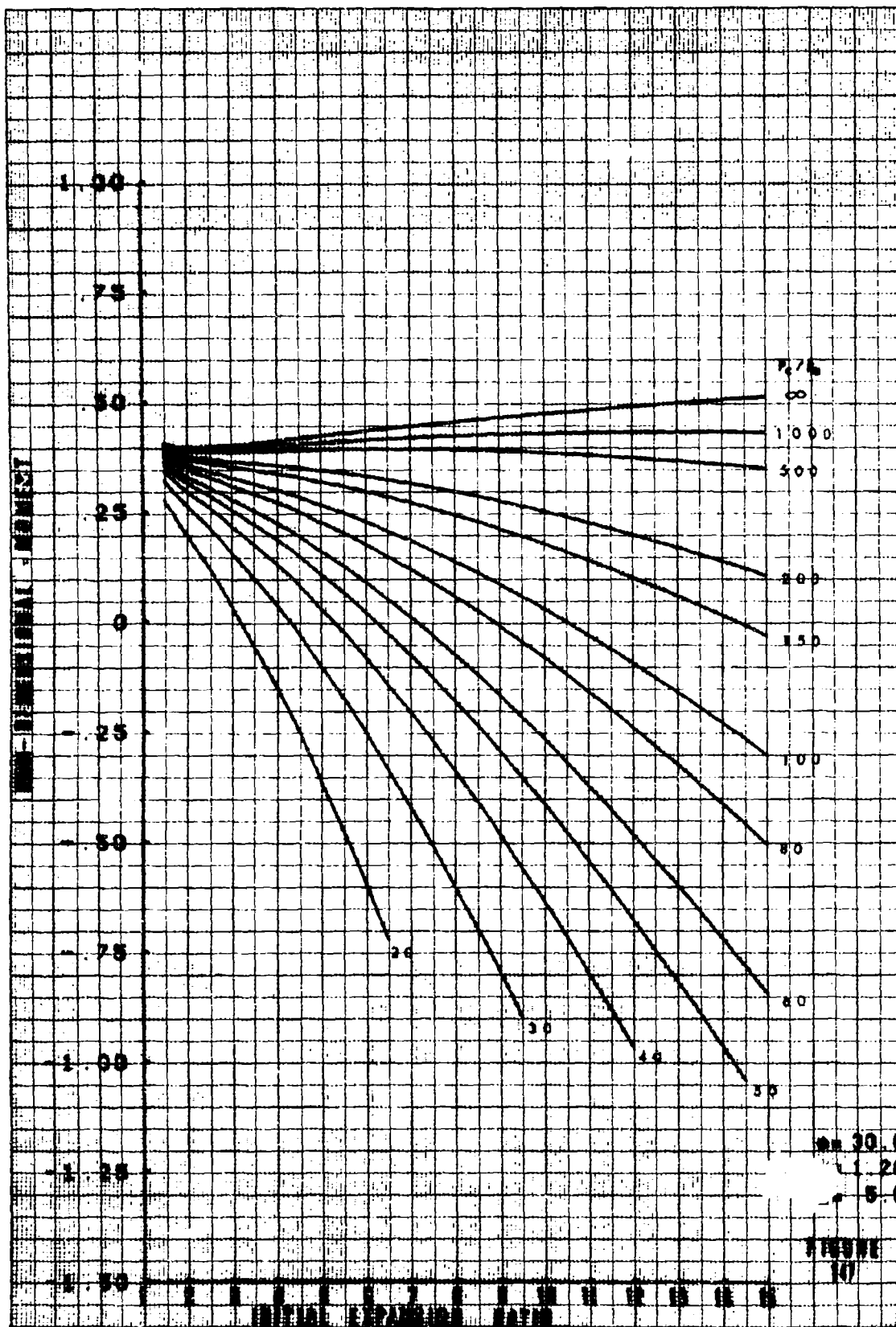






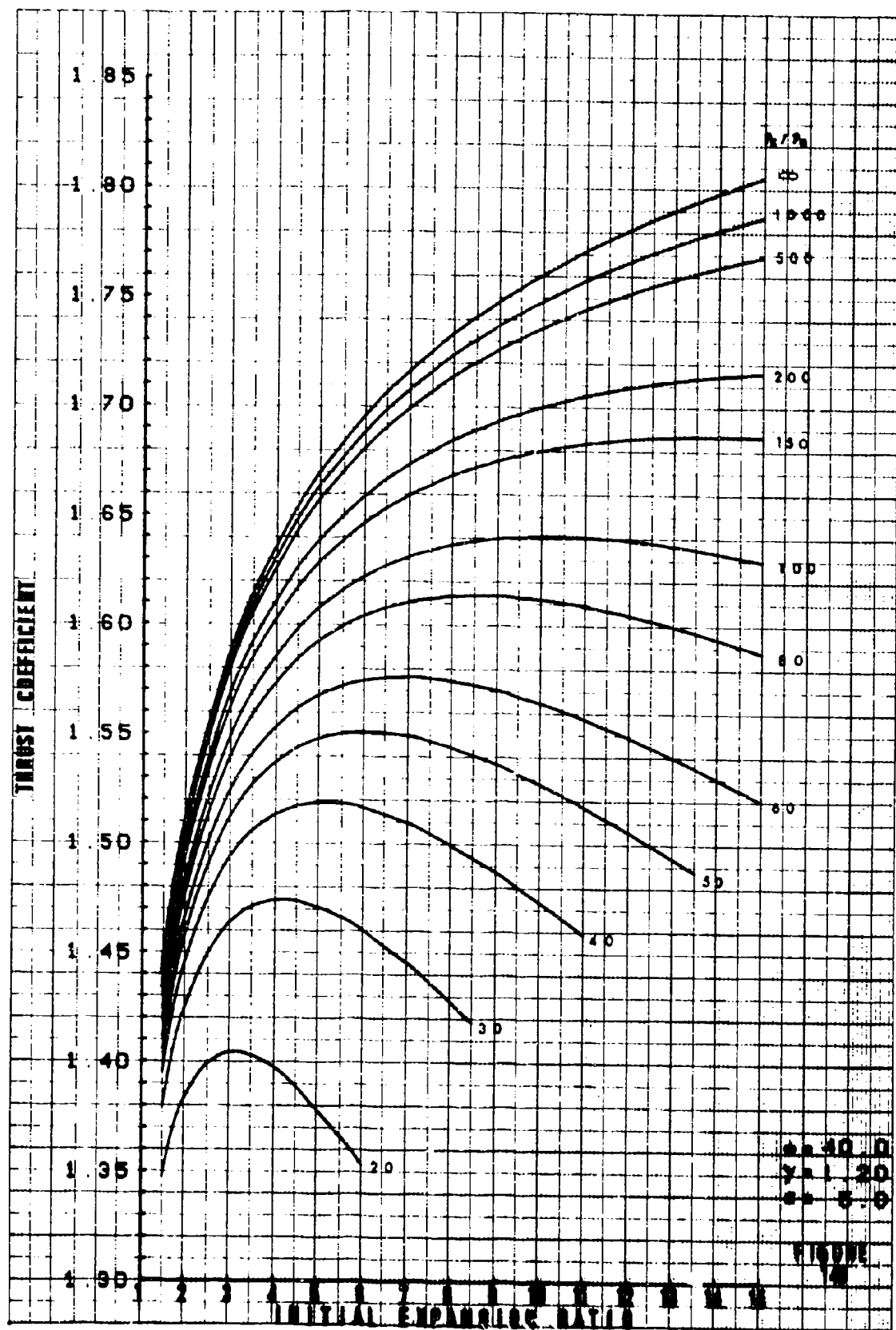


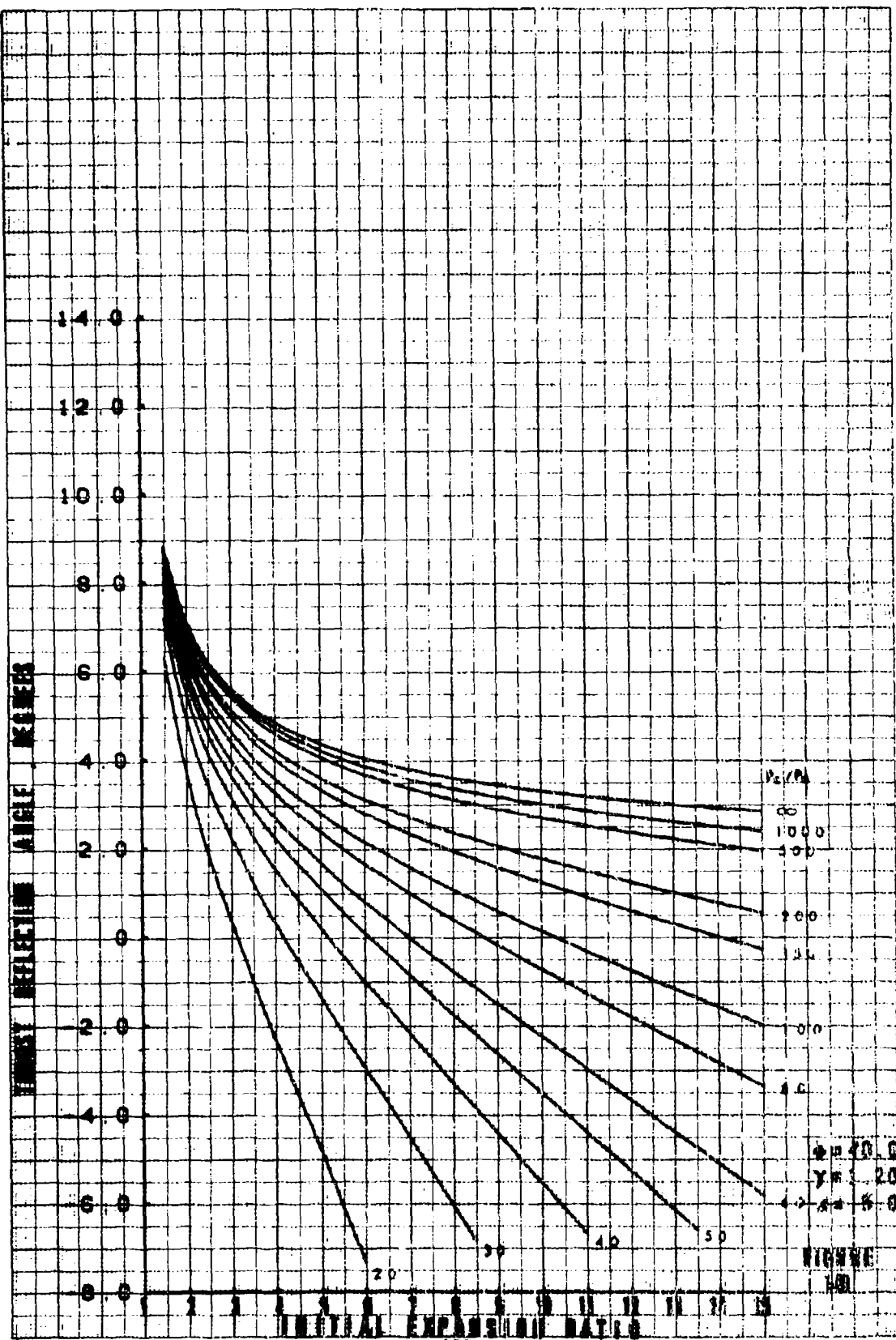




30.0
20
5.0

FIGURE 10





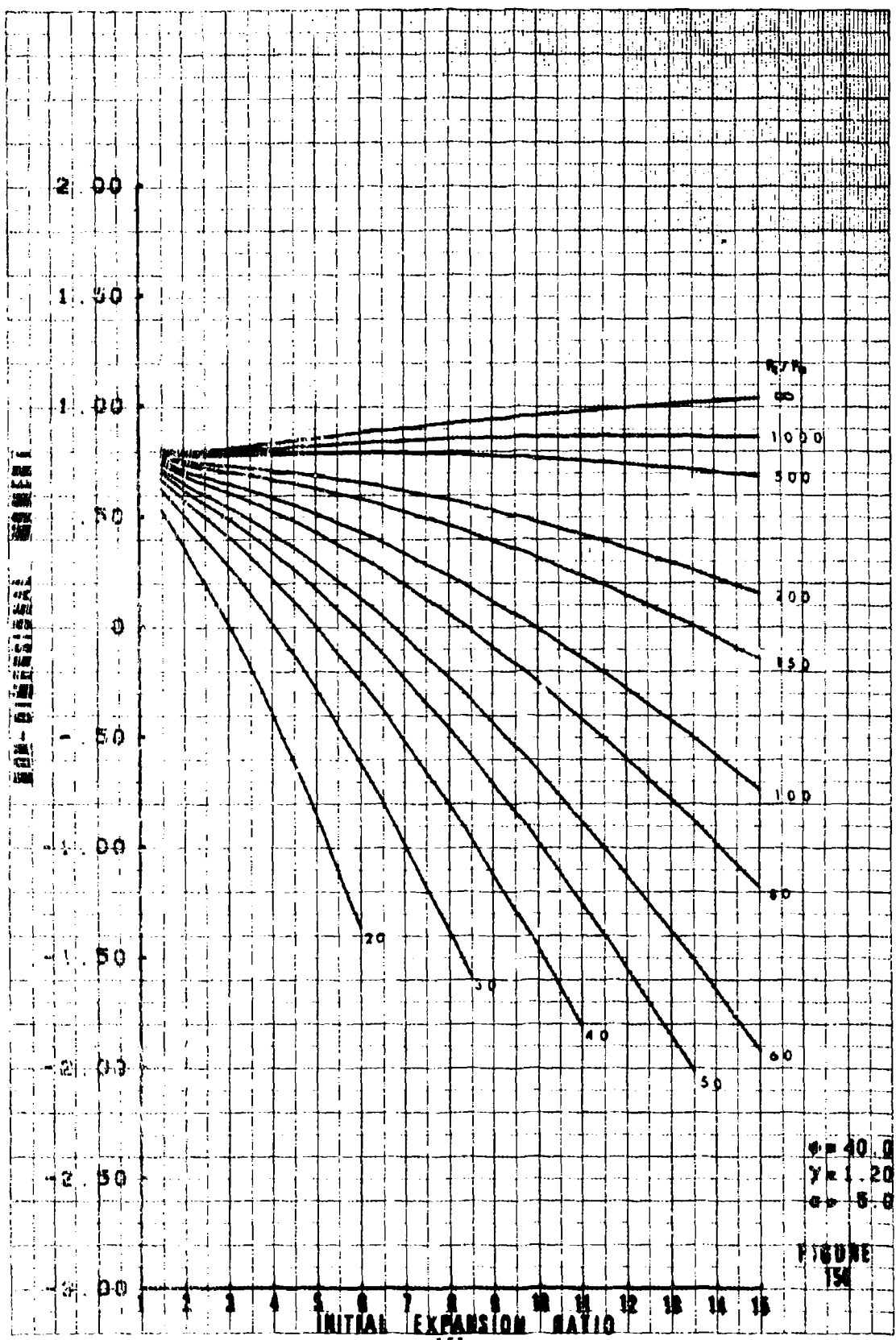
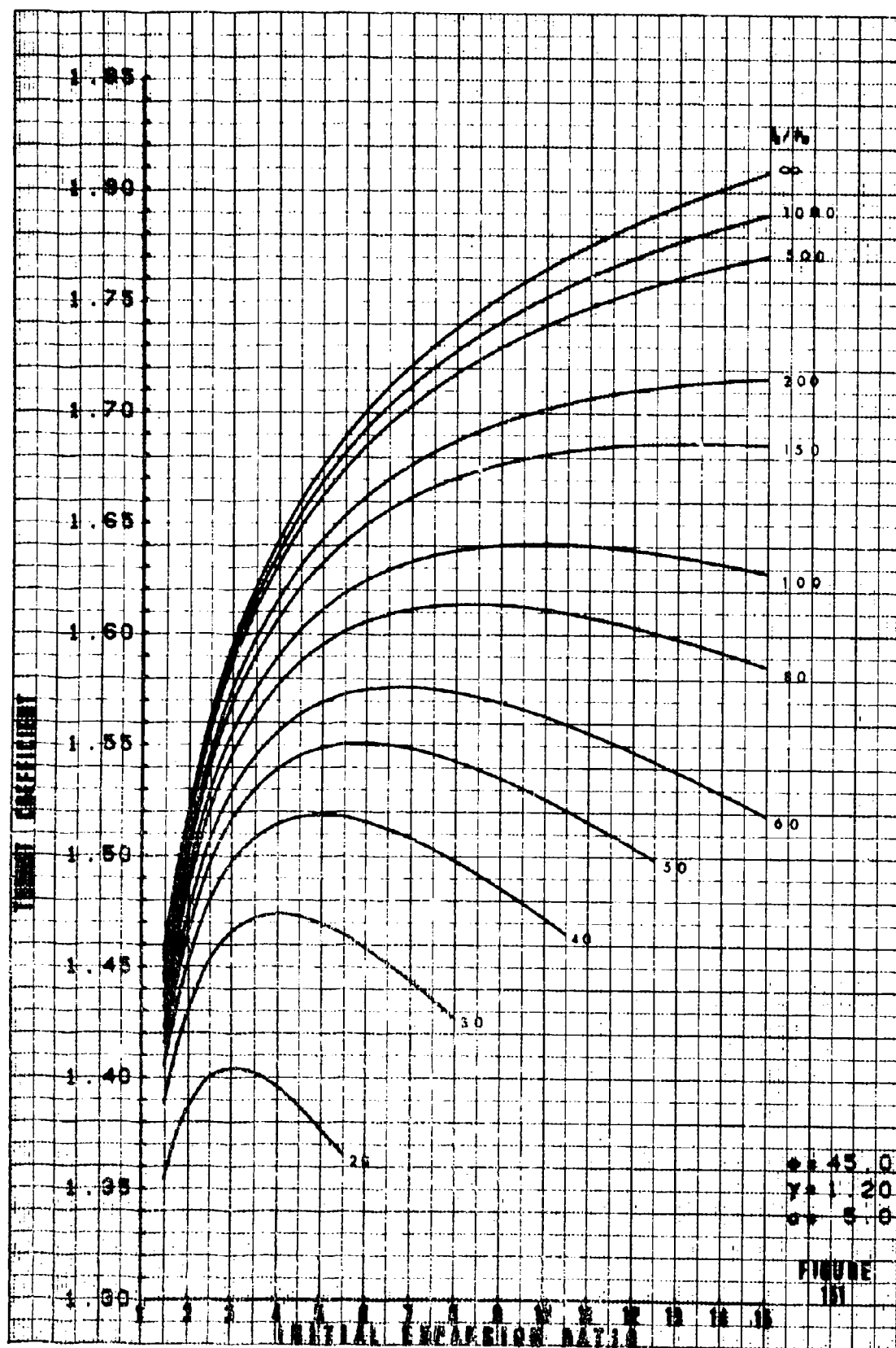


FIGURE 154



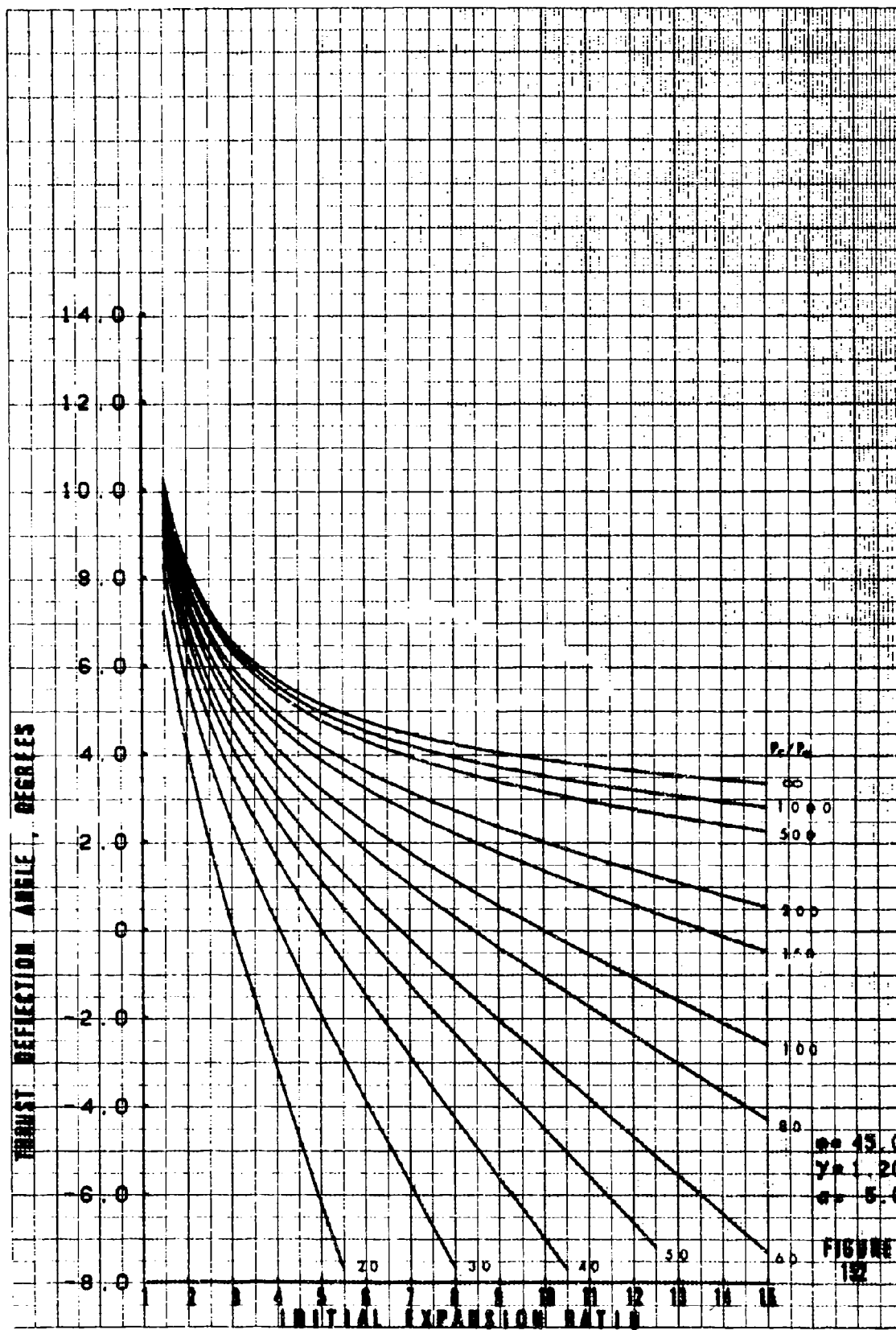
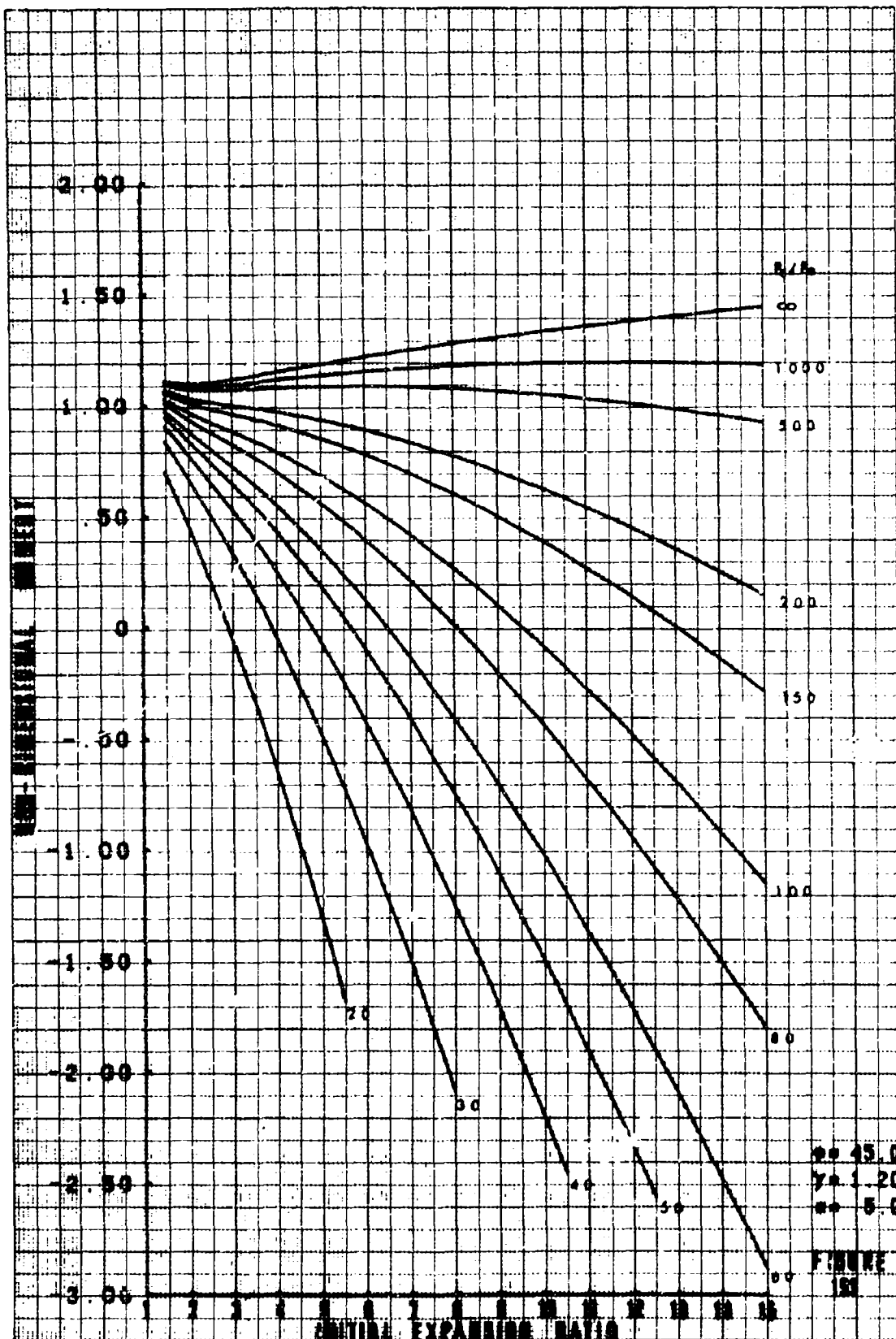


FIGURE 192



$\mu = 45.0$
 $\gamma = 1.20$
 $\sigma = 5.0$

FIGURE 154

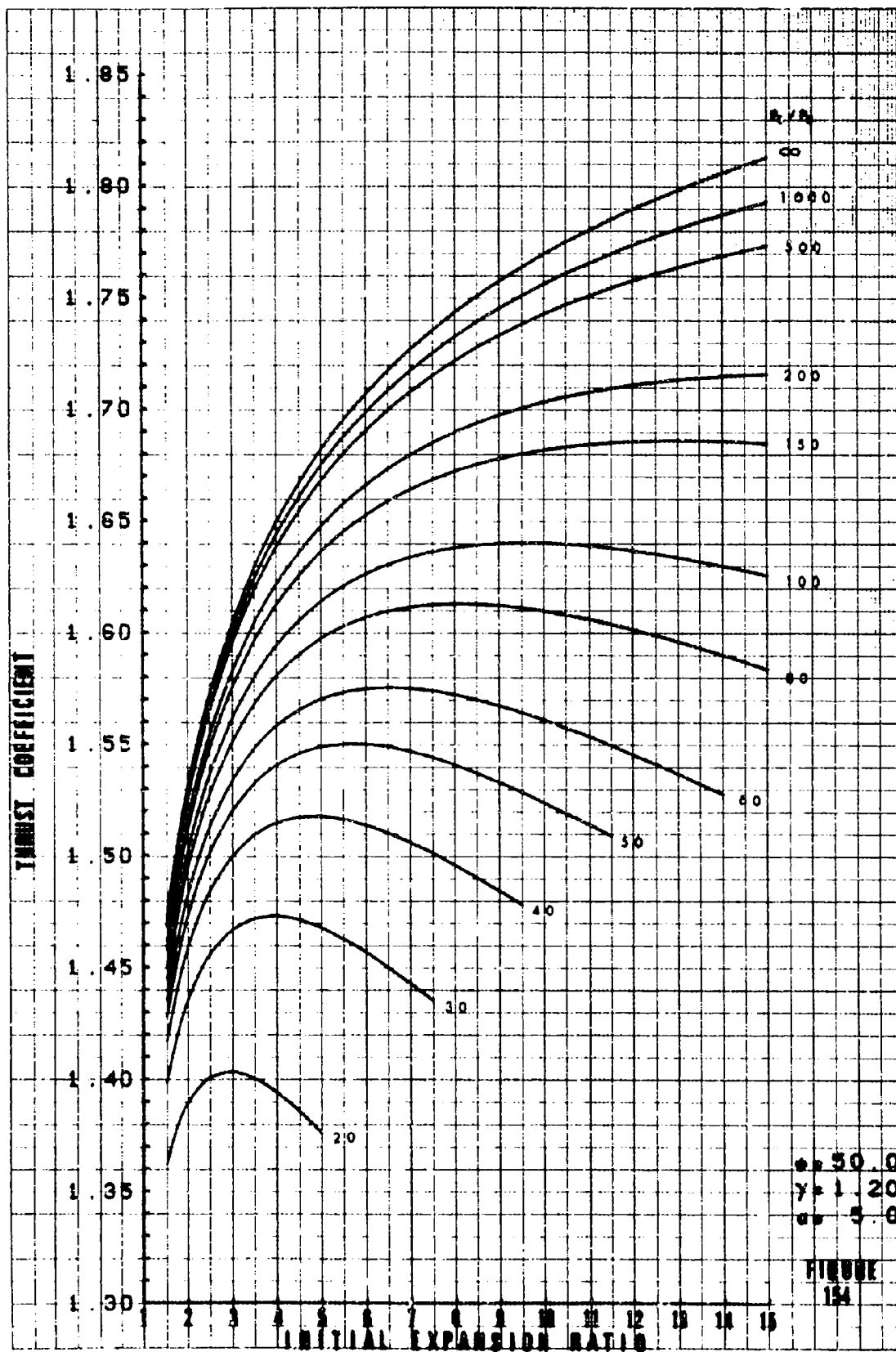
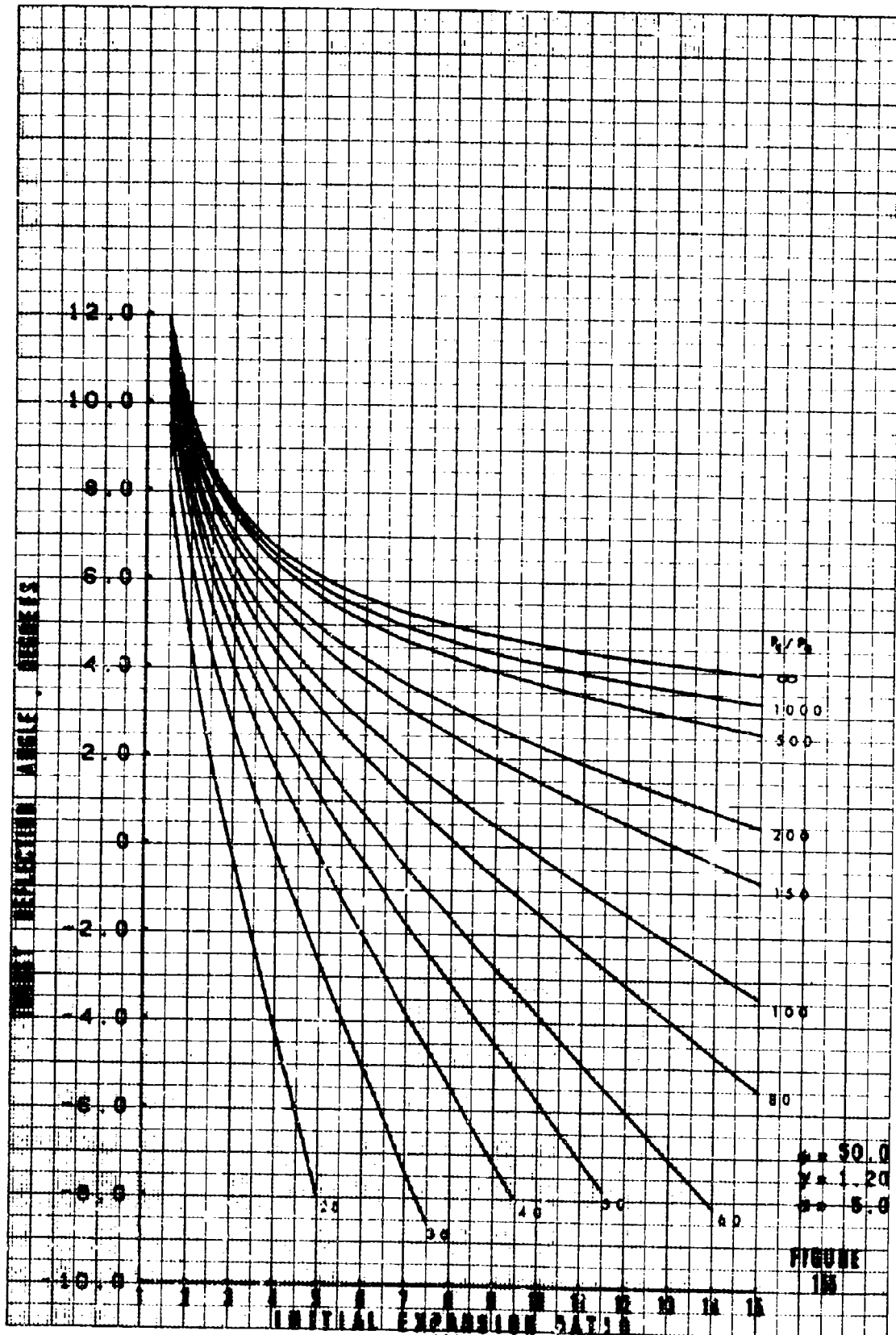
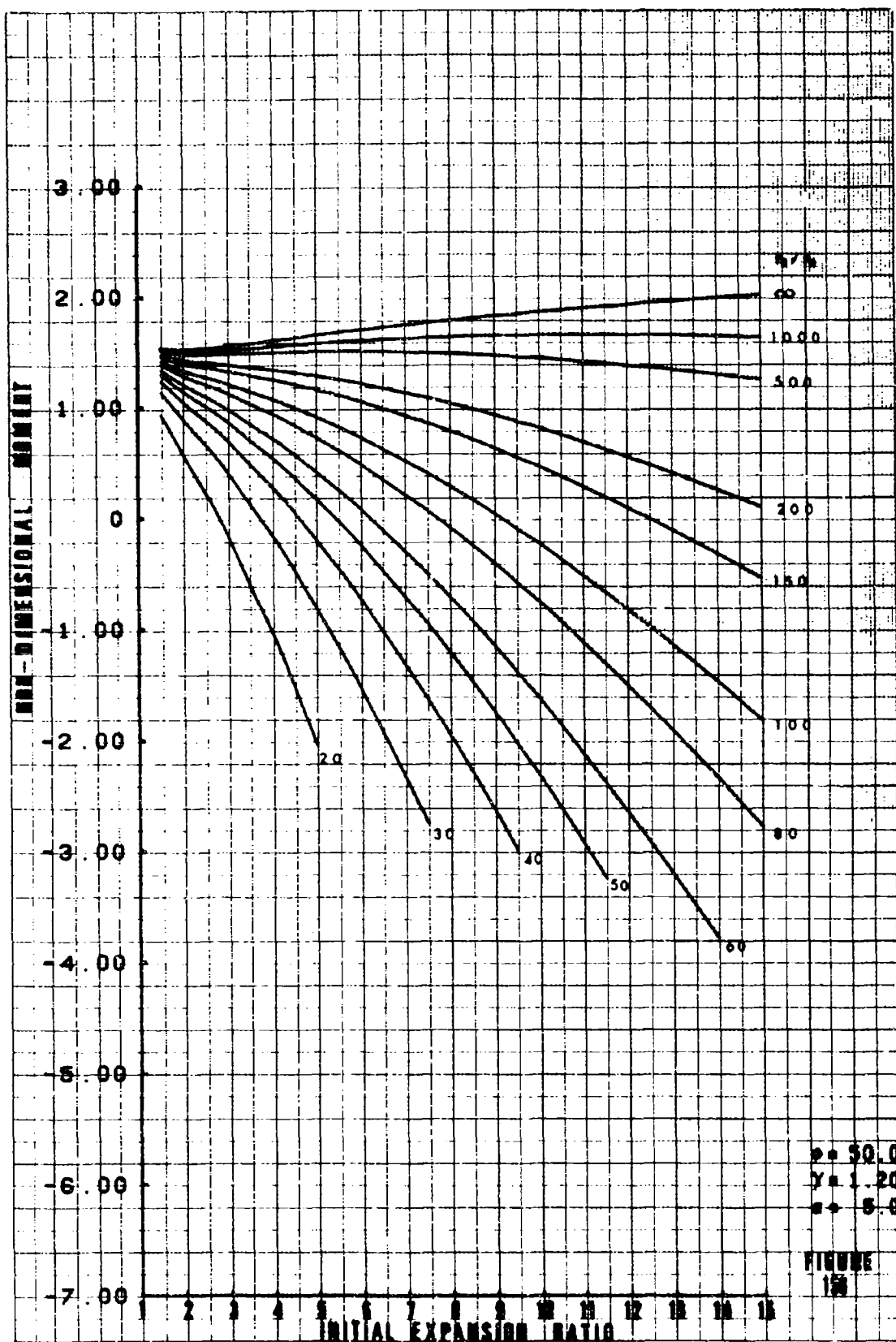
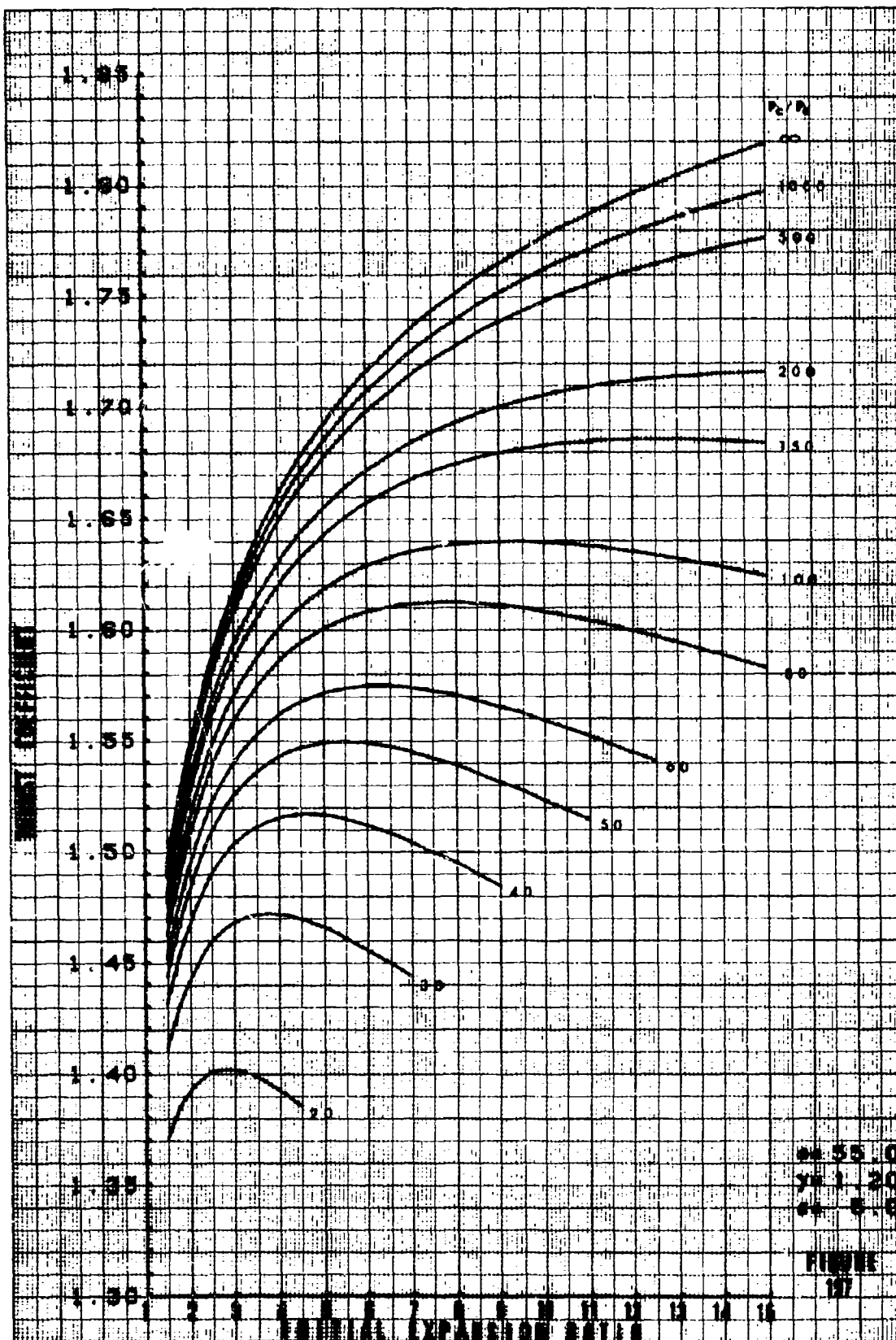
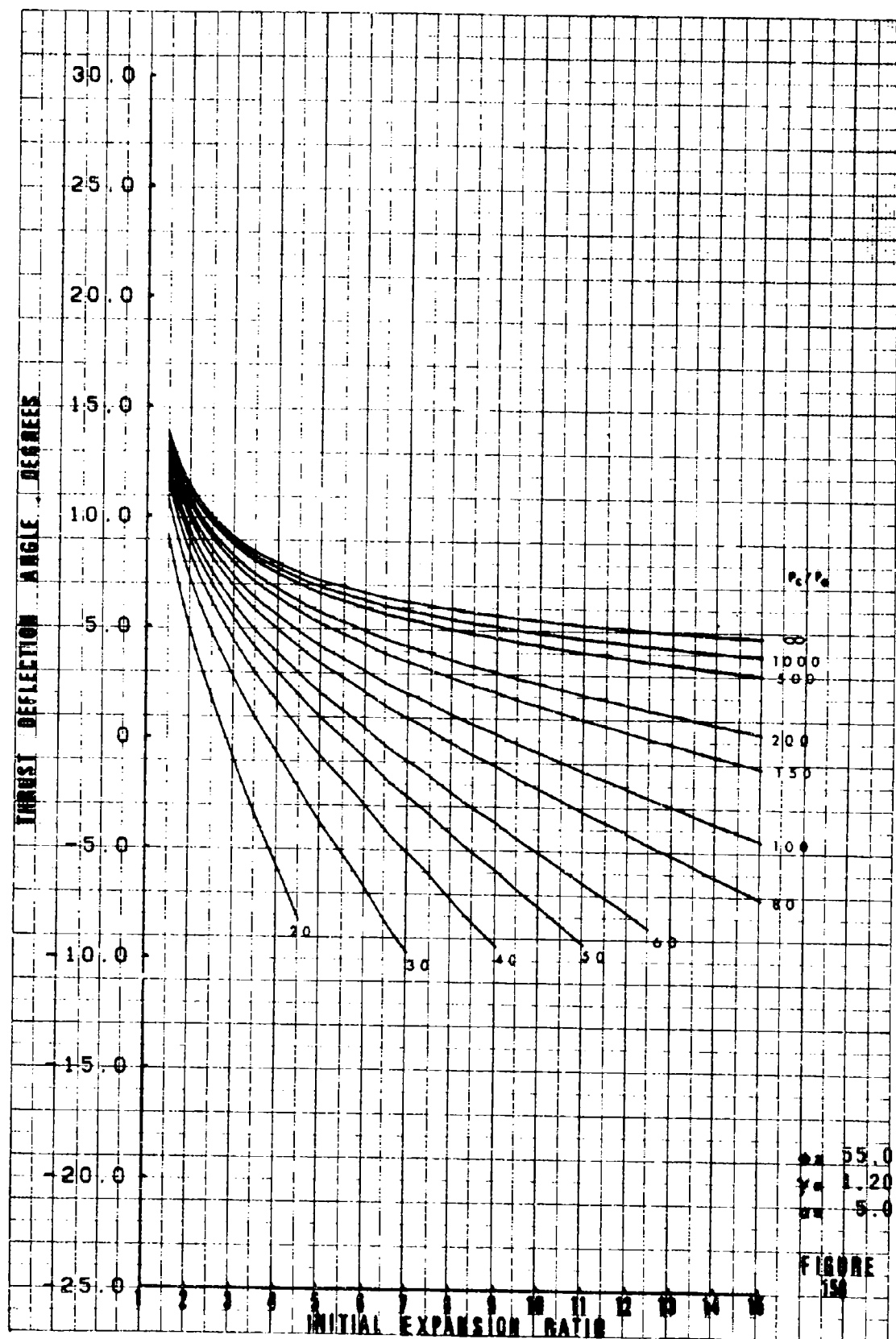


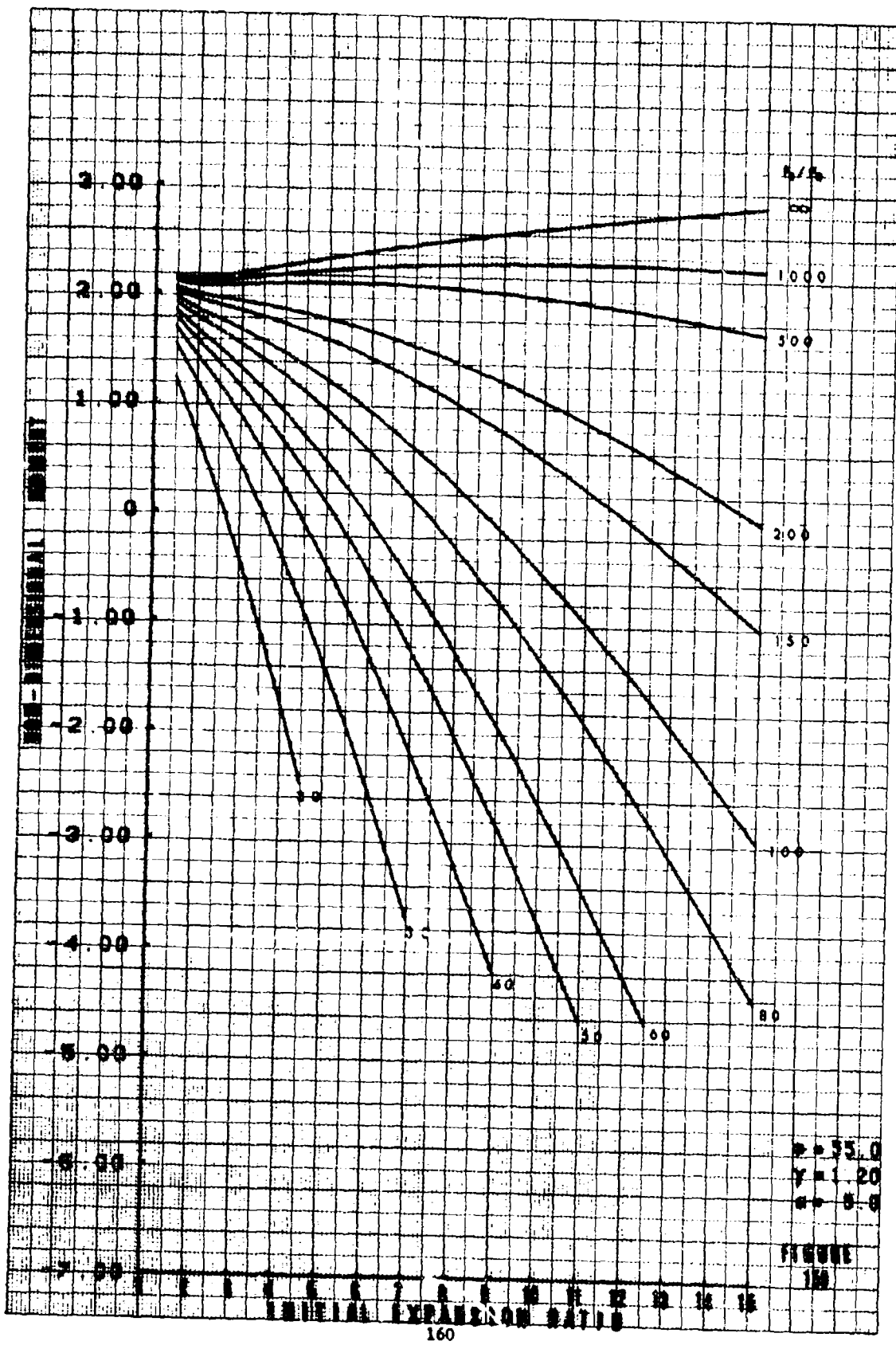
FIGURE
154

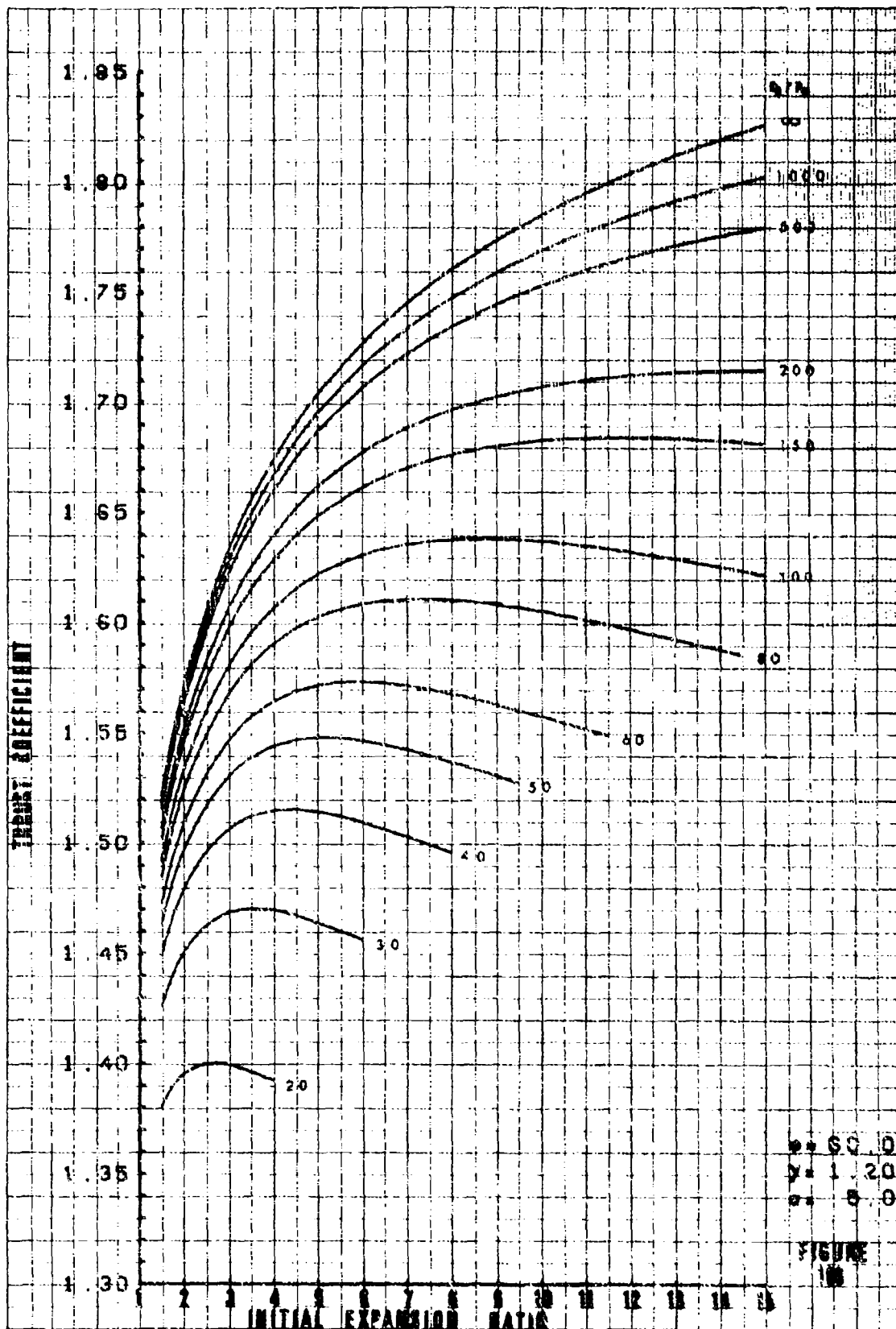


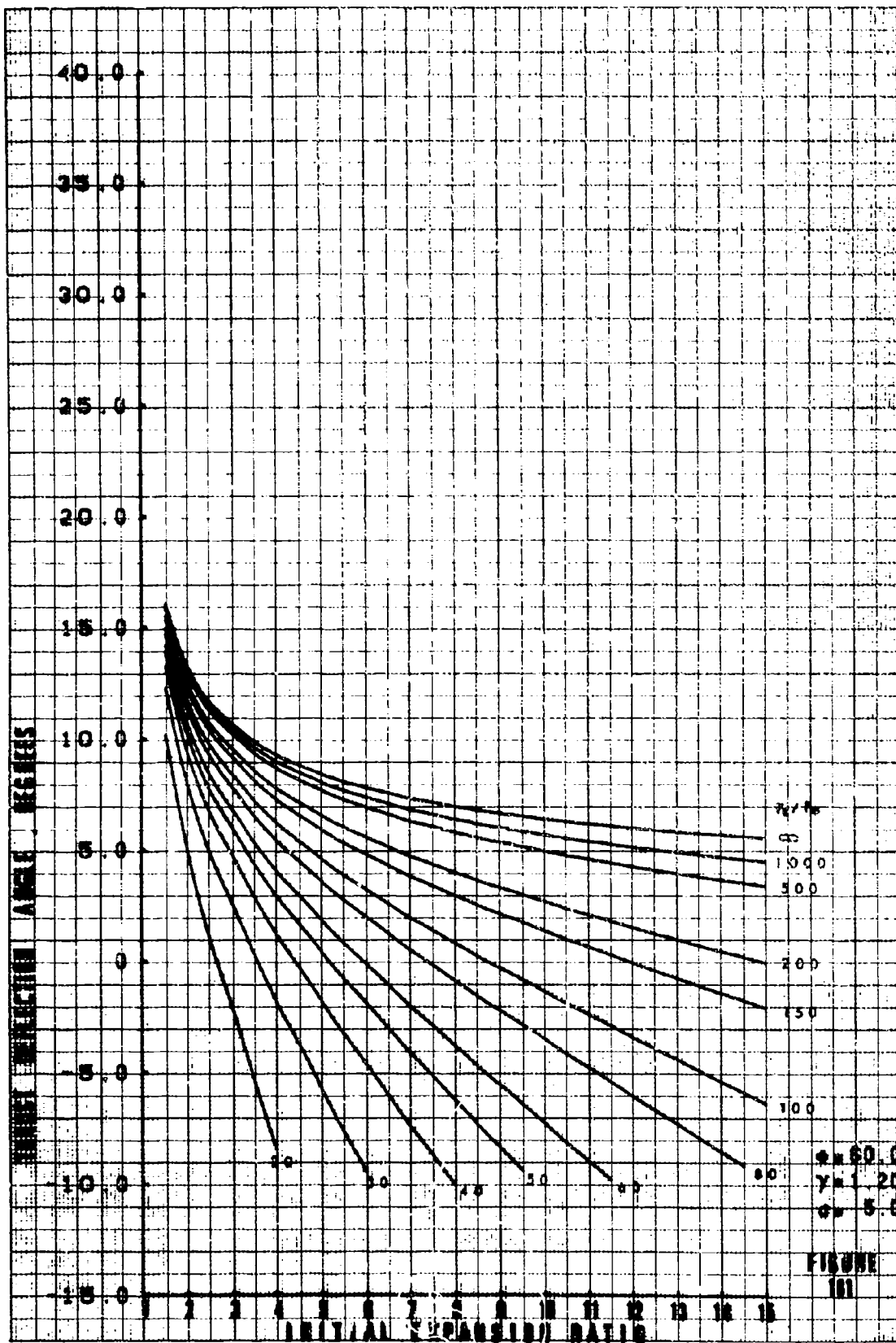


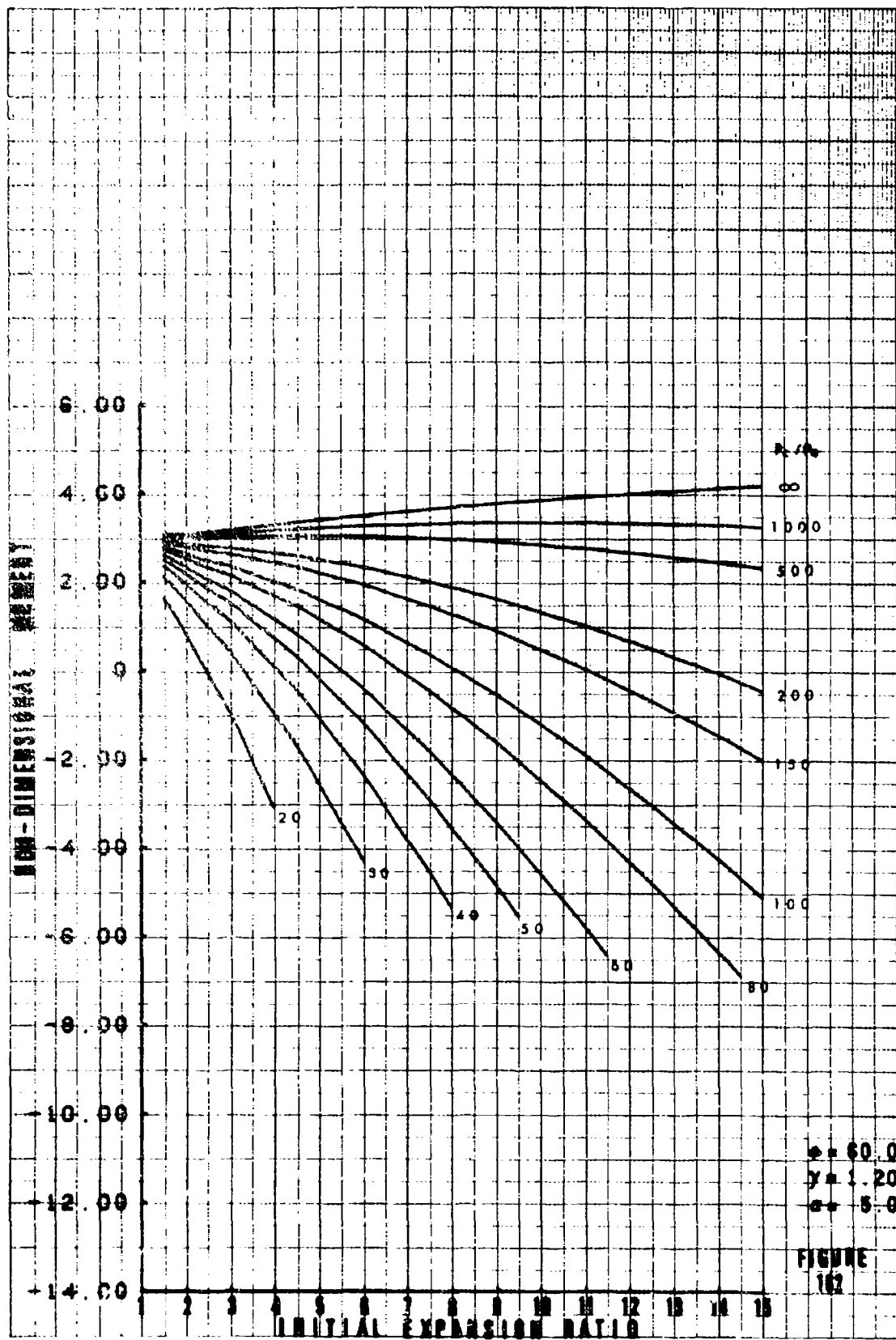


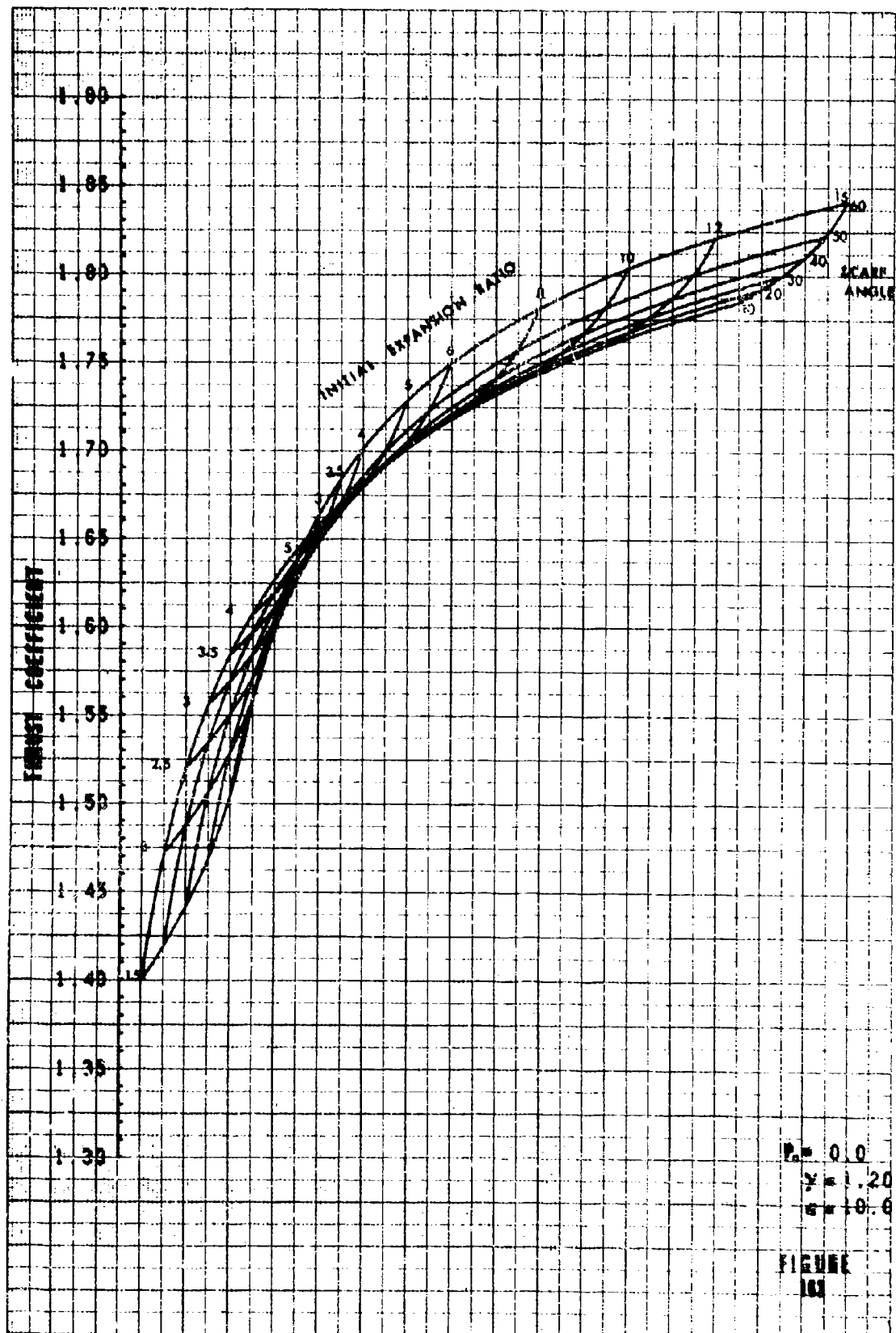


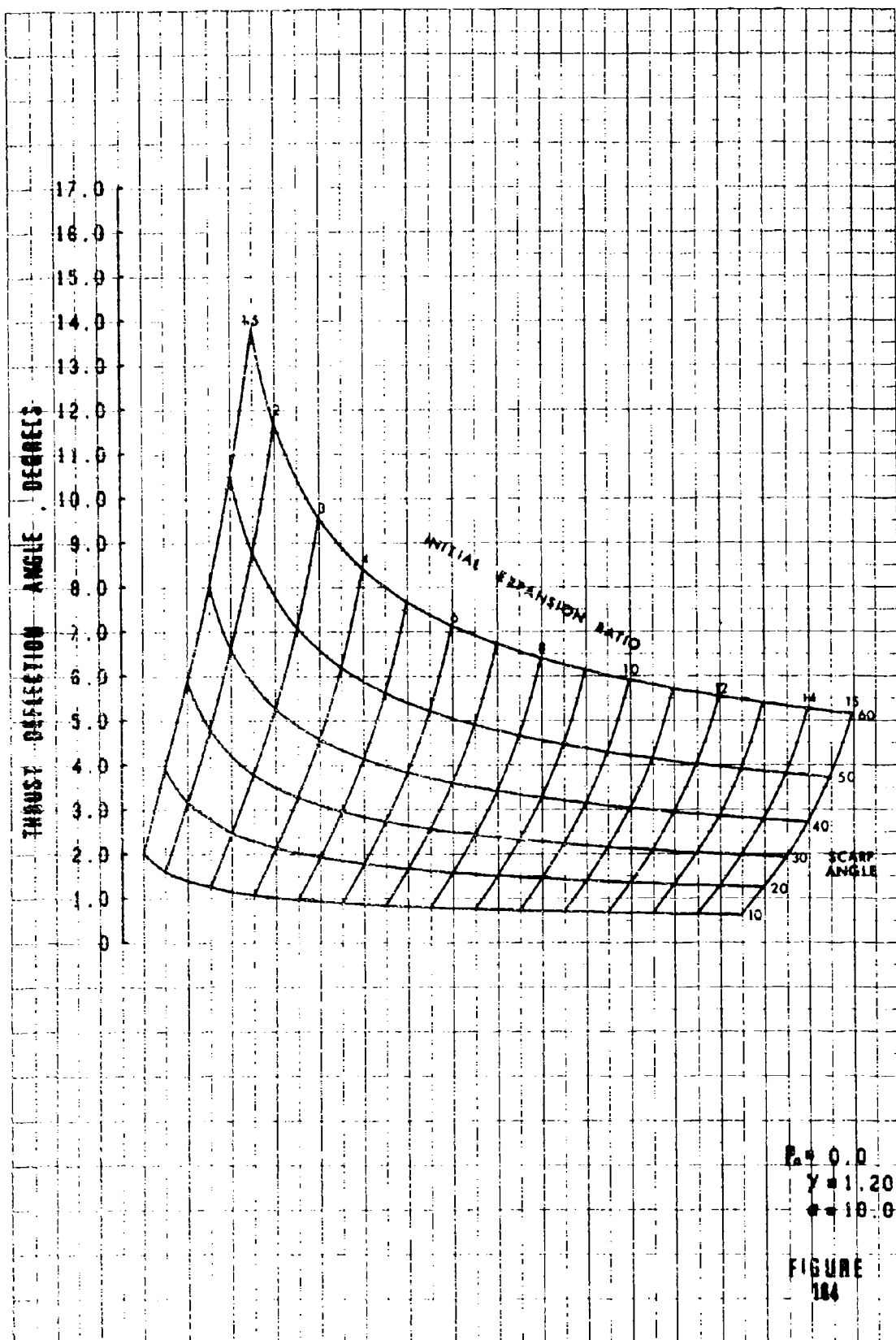


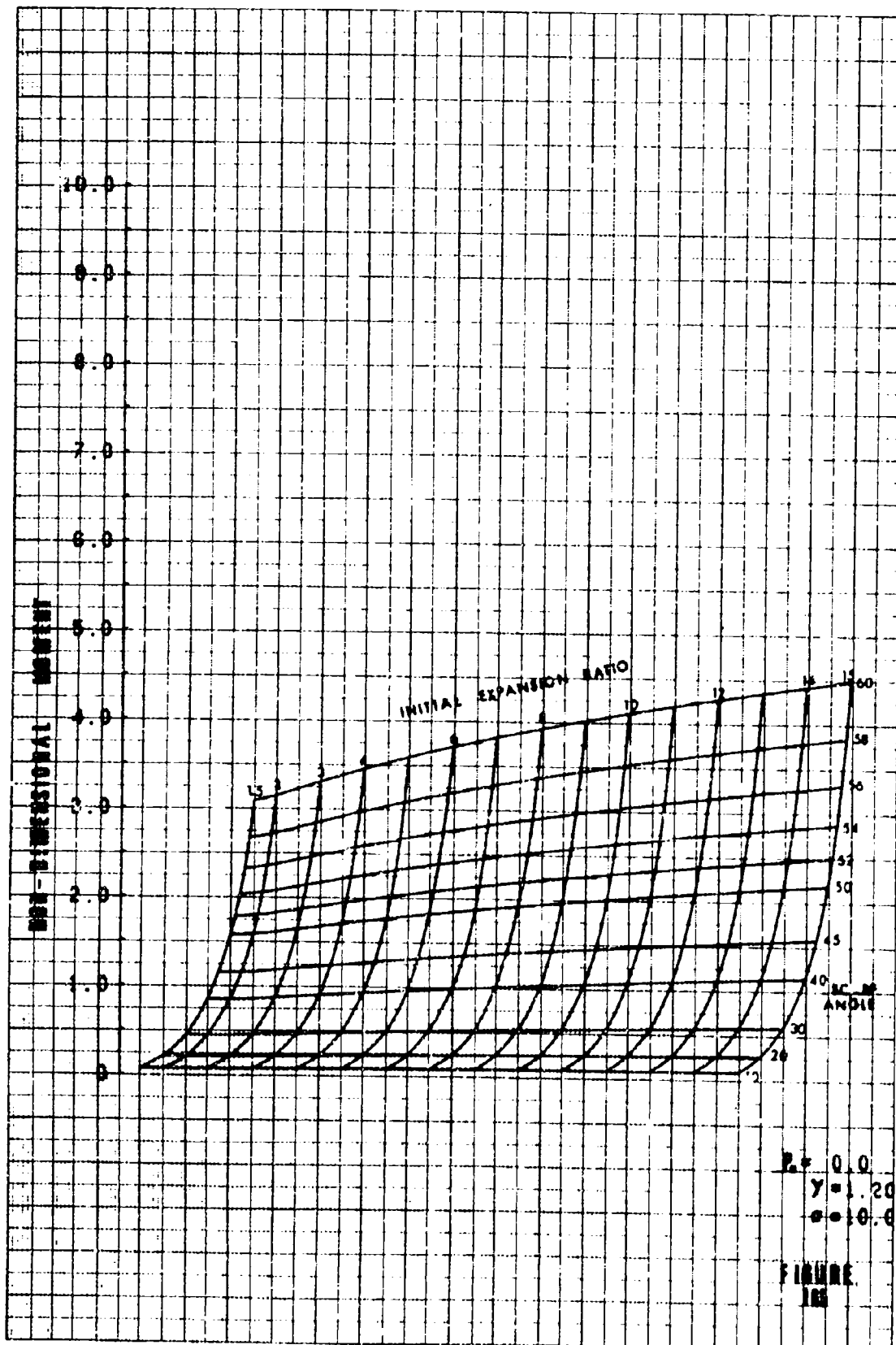


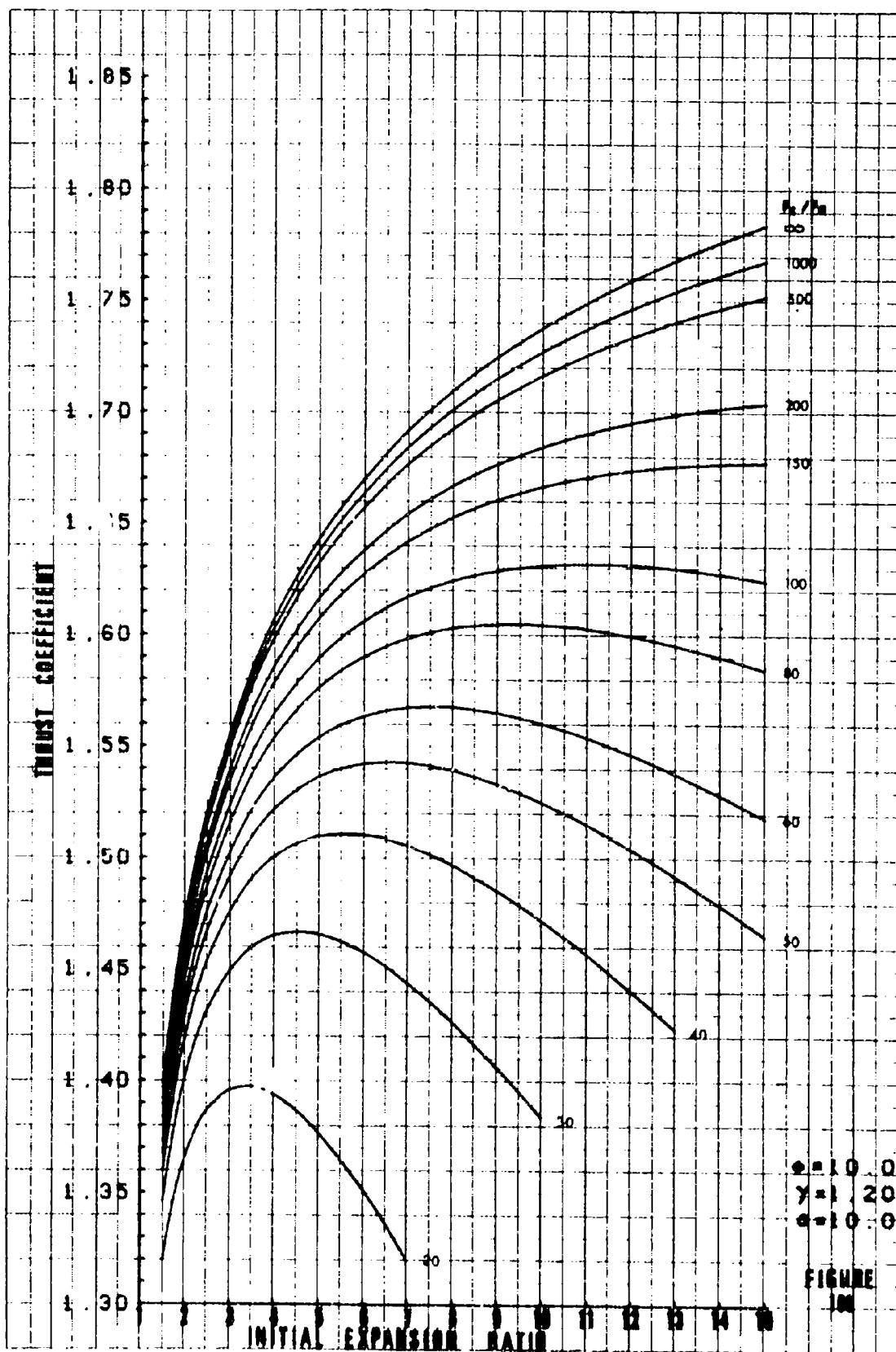






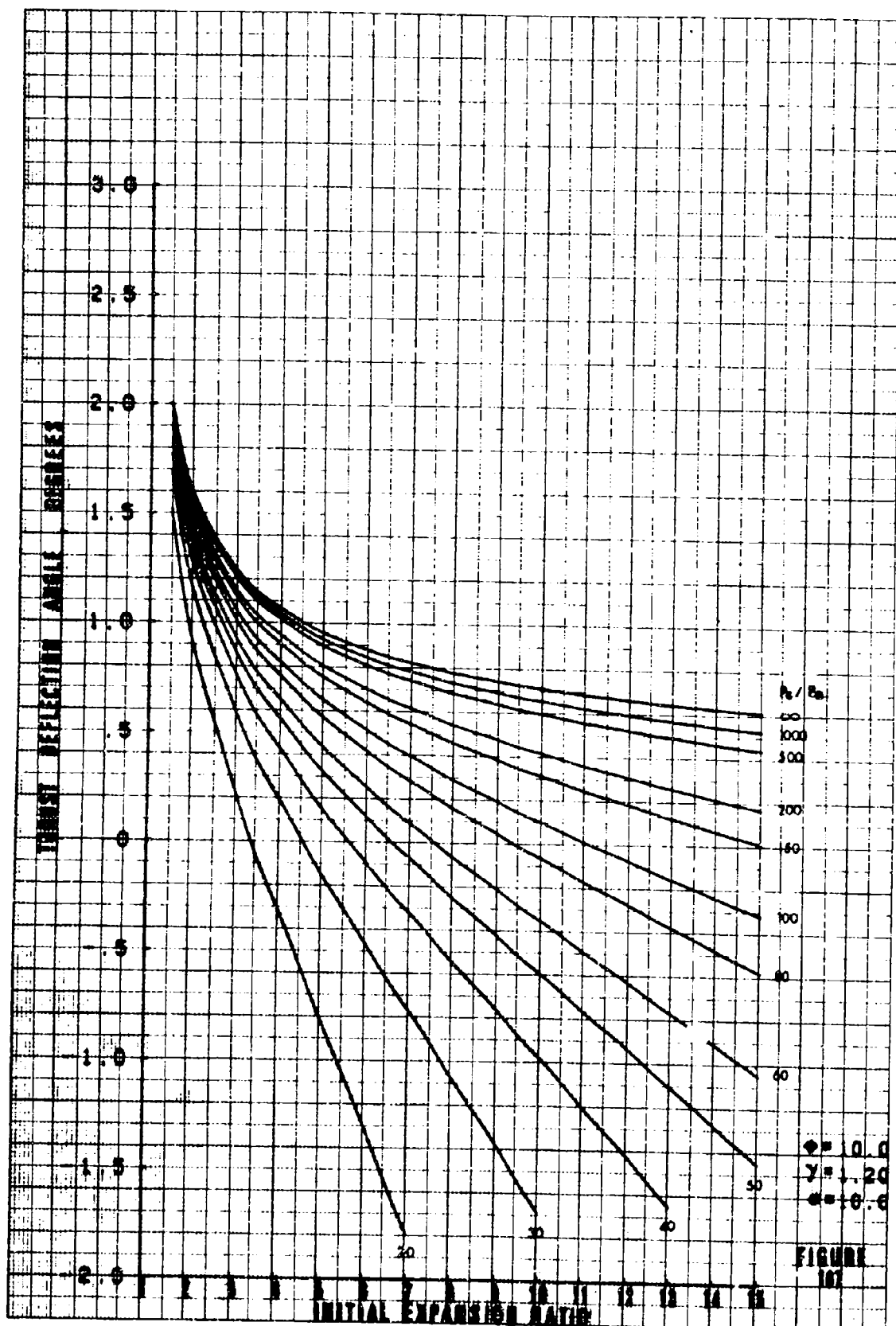


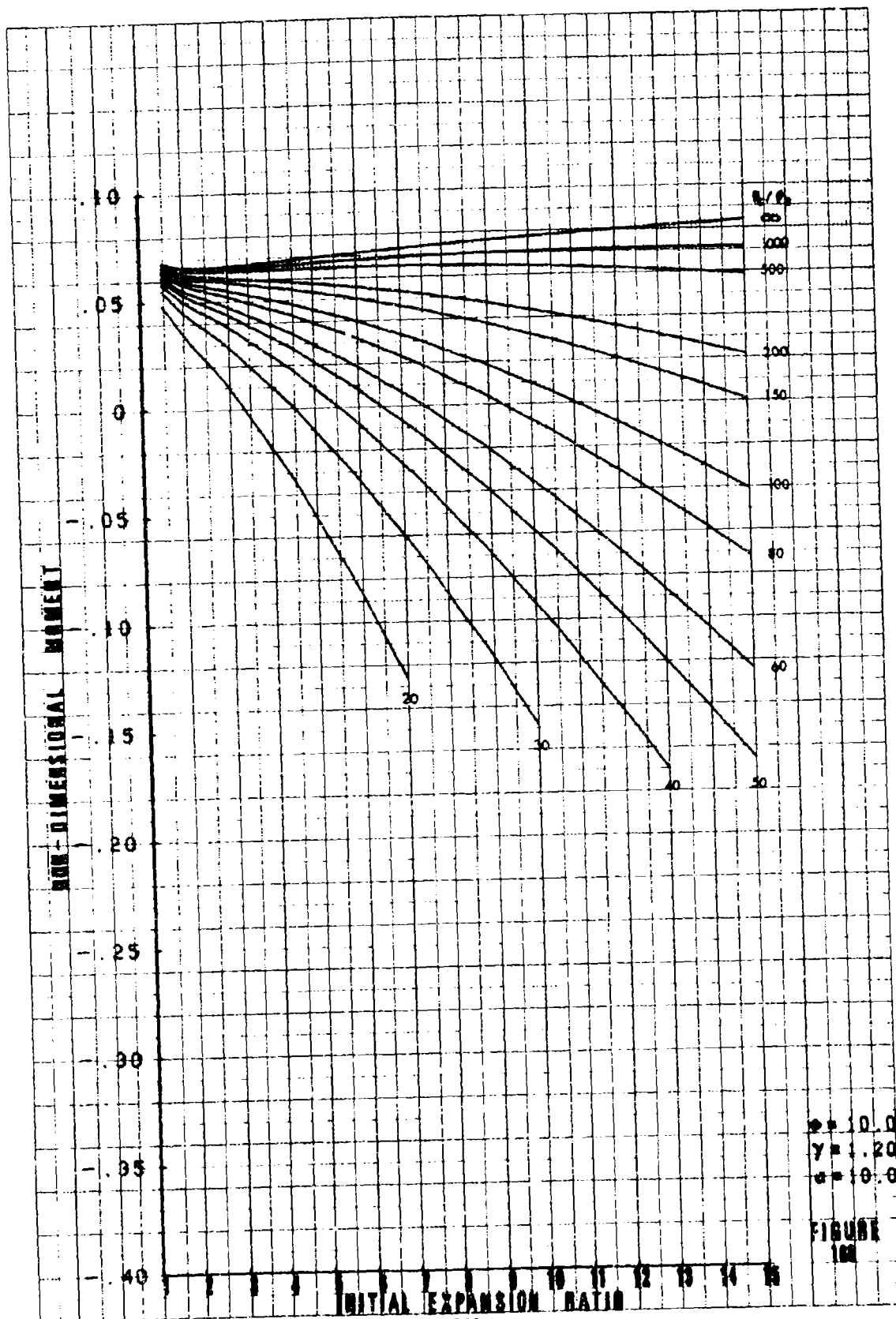


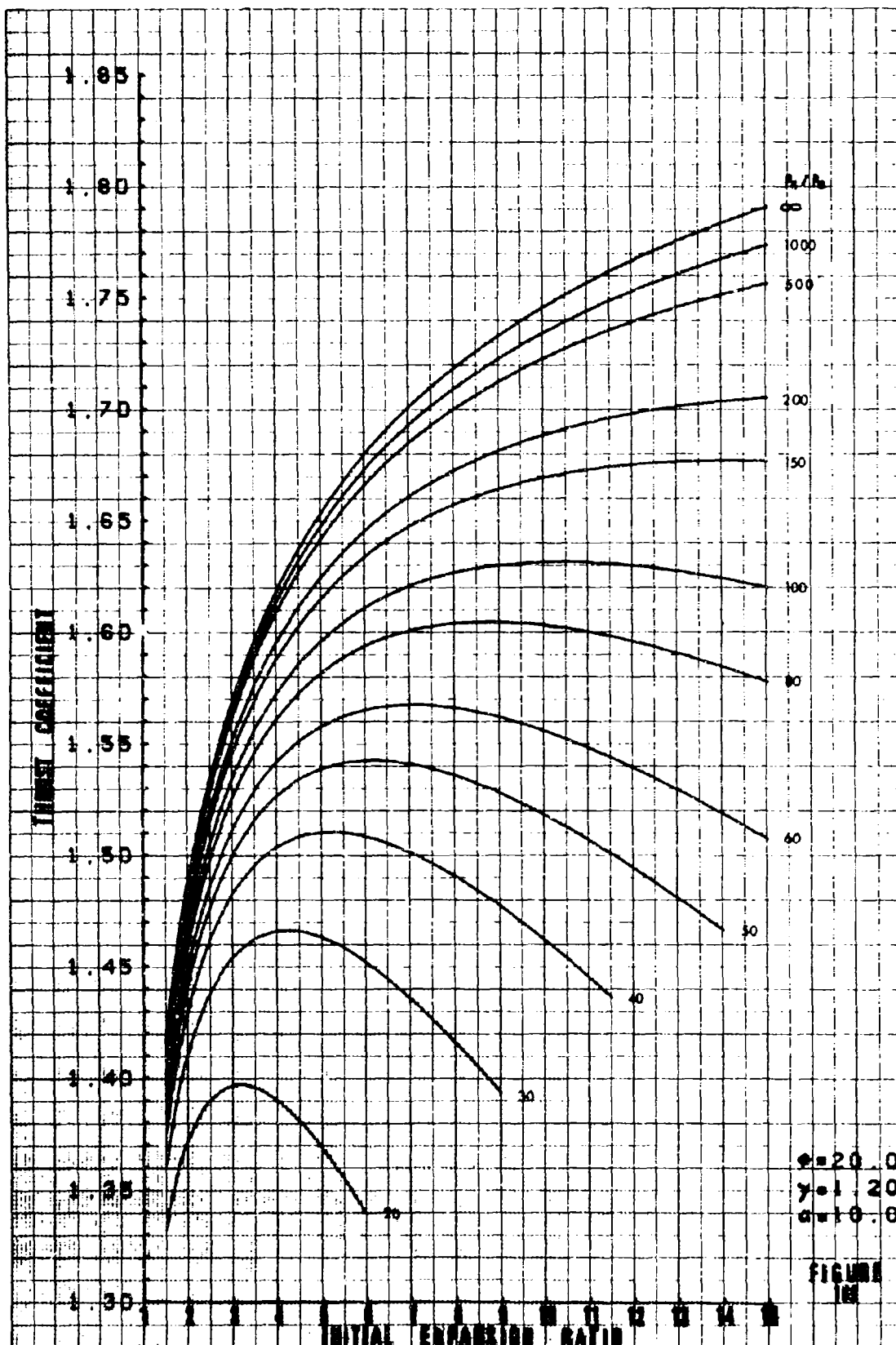


$\phi = 10.0$
 $\gamma = 1.20$
 $\alpha = 10.0$

FIGURE 100

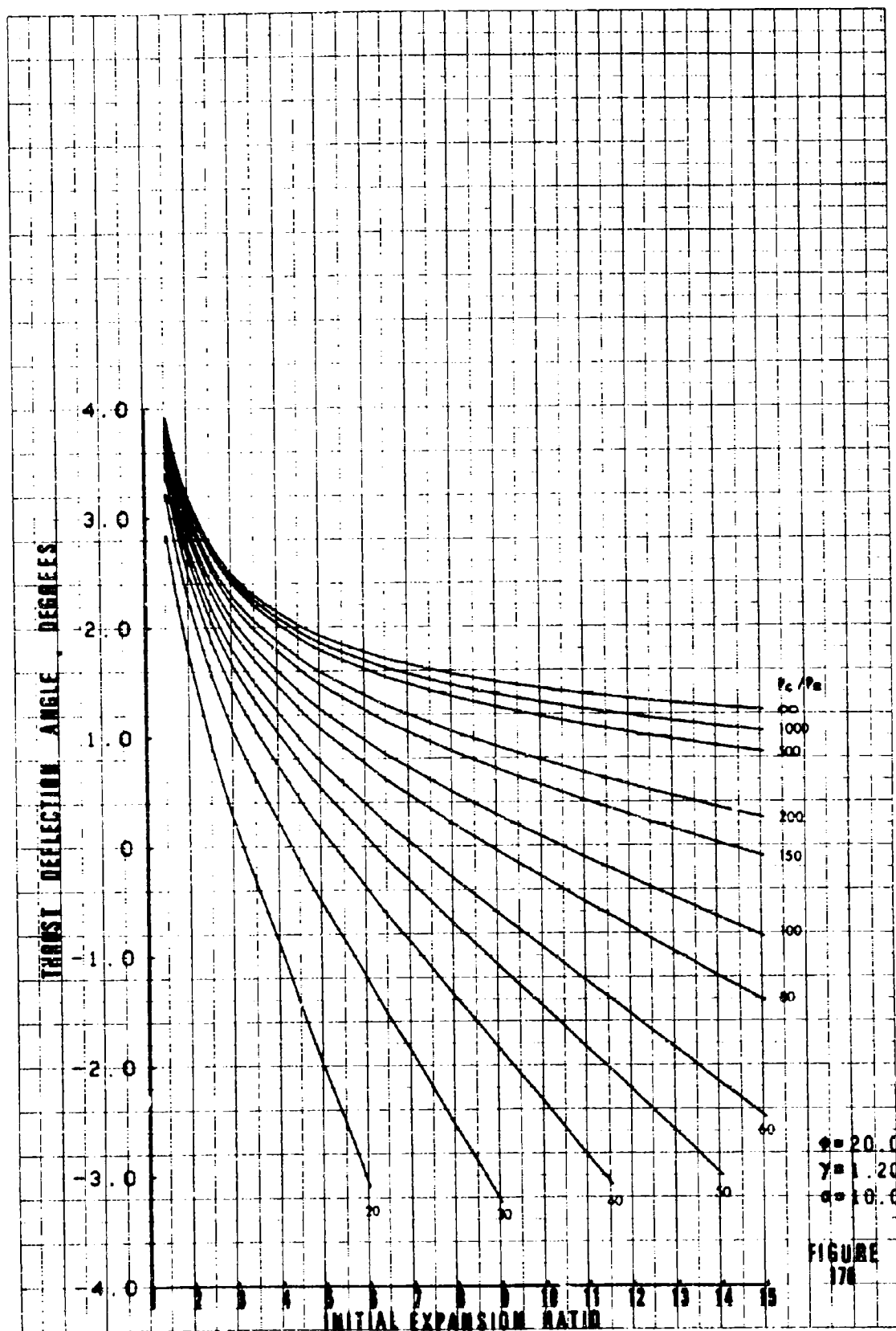


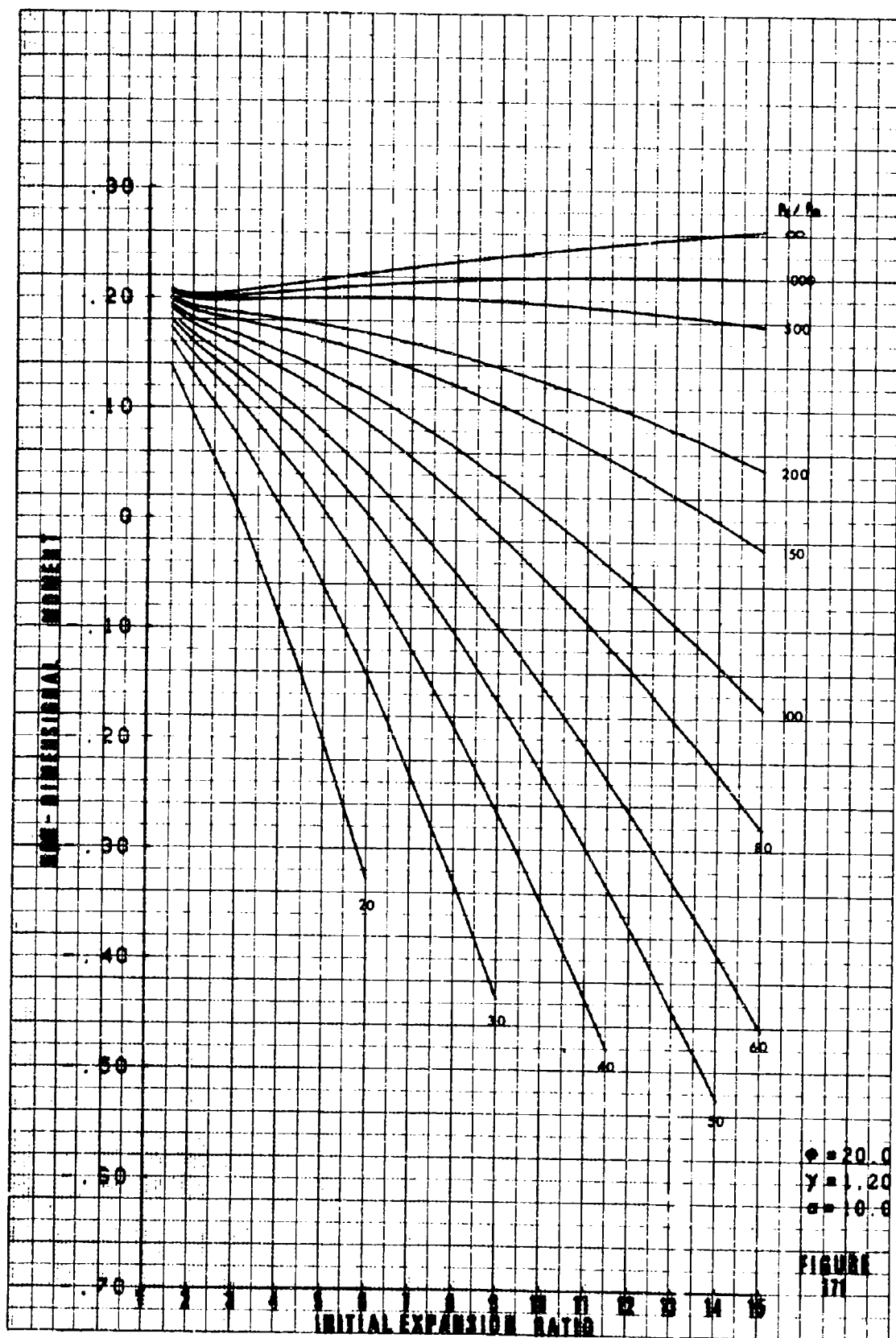


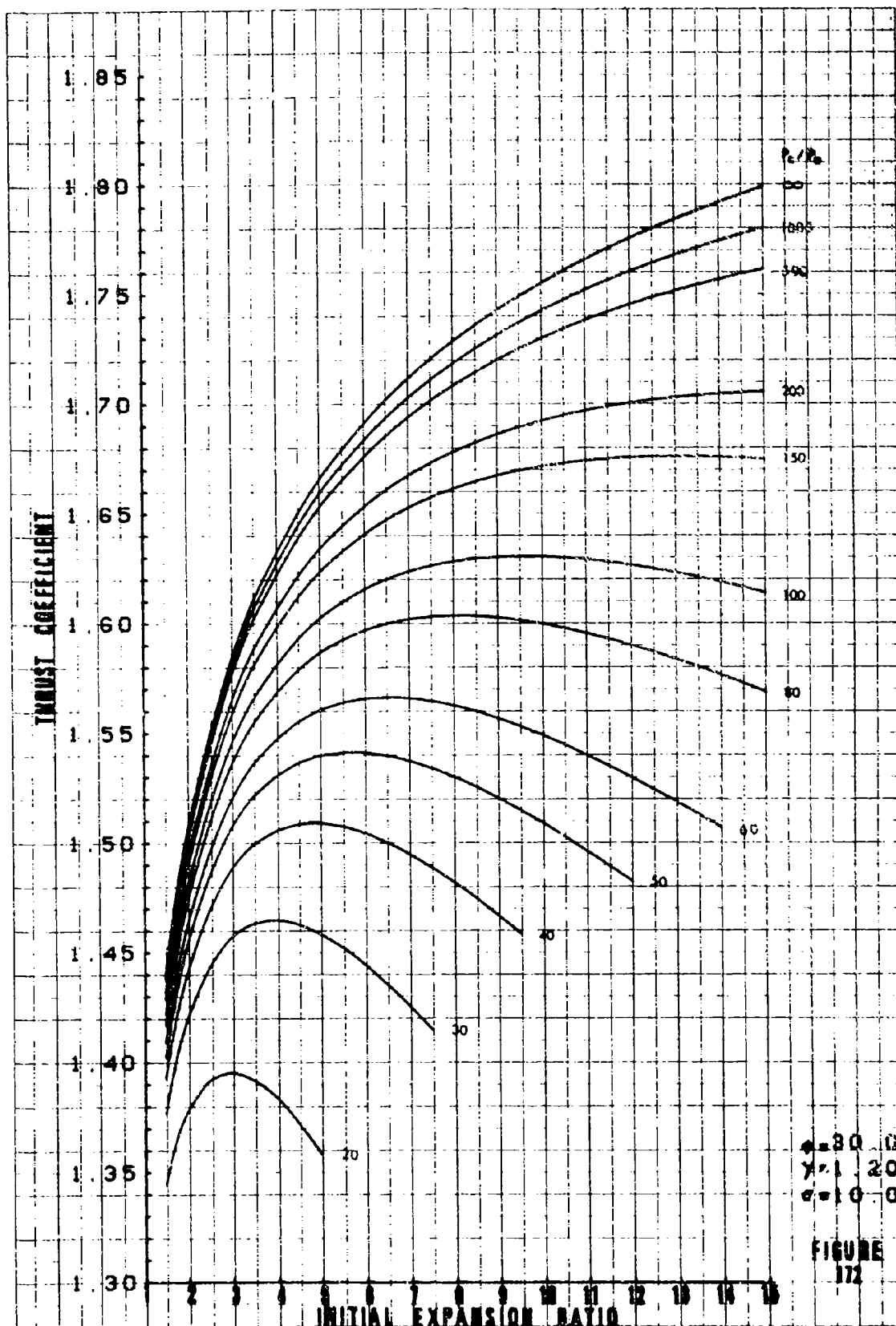


$\phi = 20.0$
 $\gamma = 1.20$
 $\alpha = 10.0$

FIGURE 10

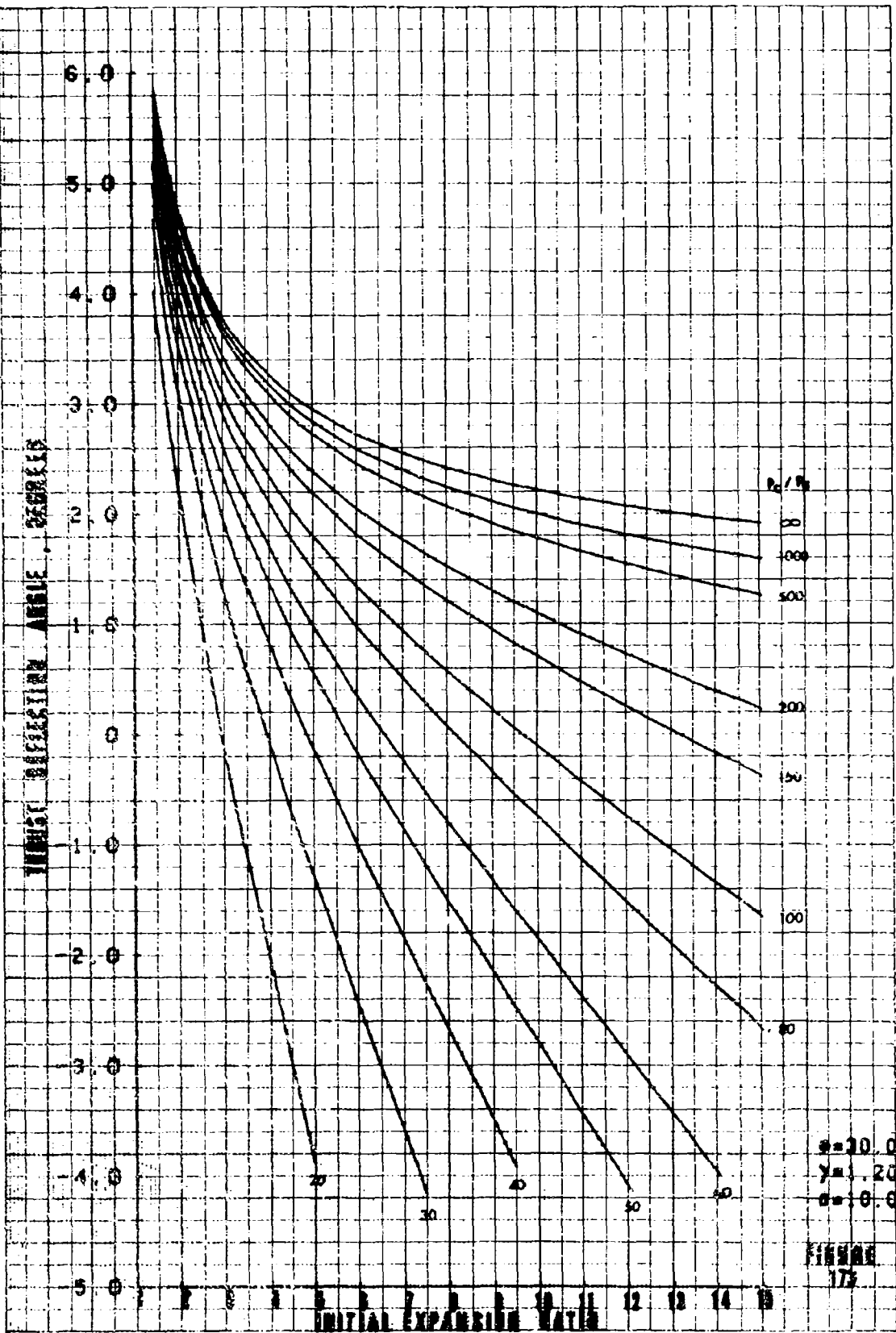


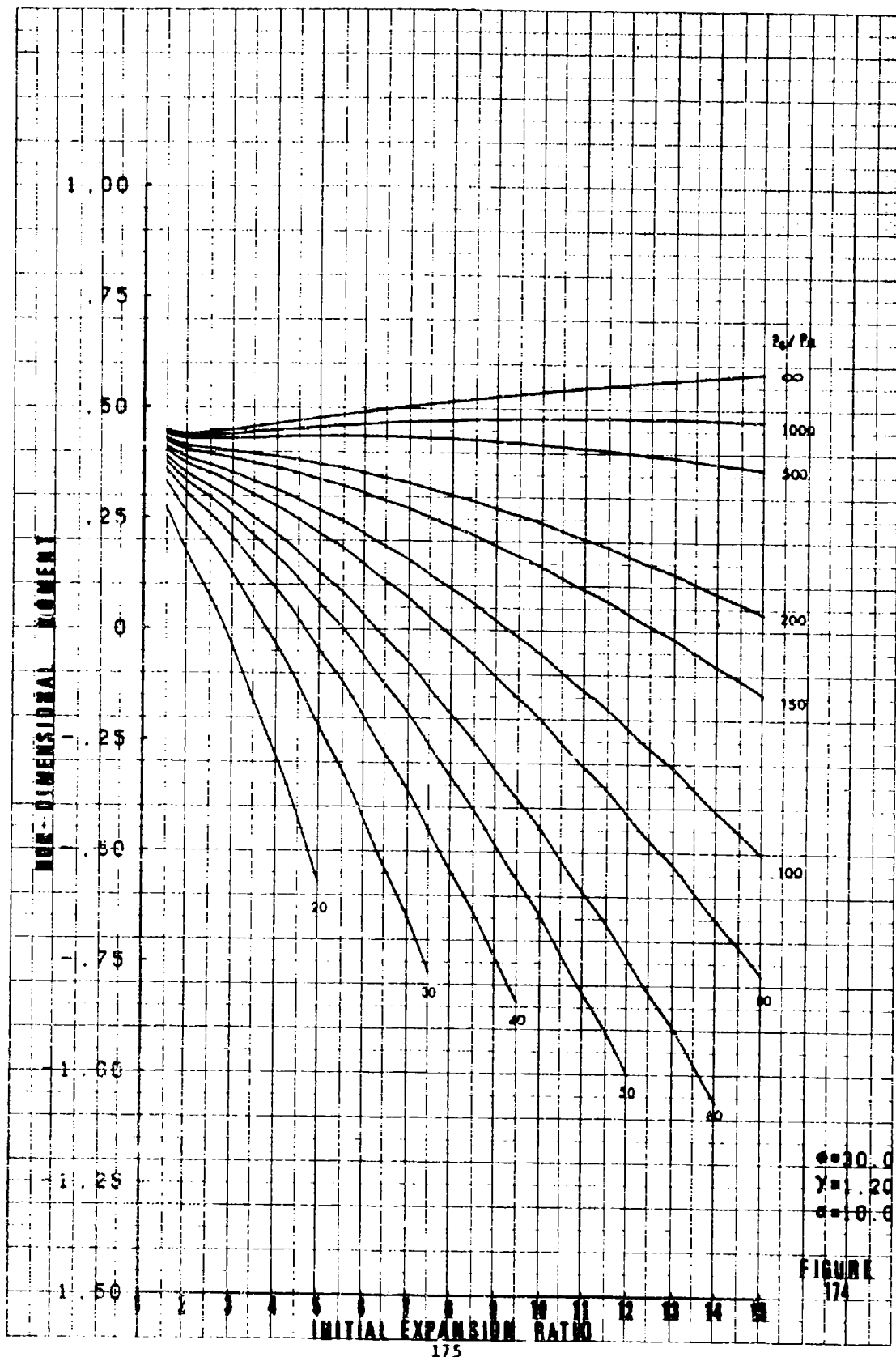


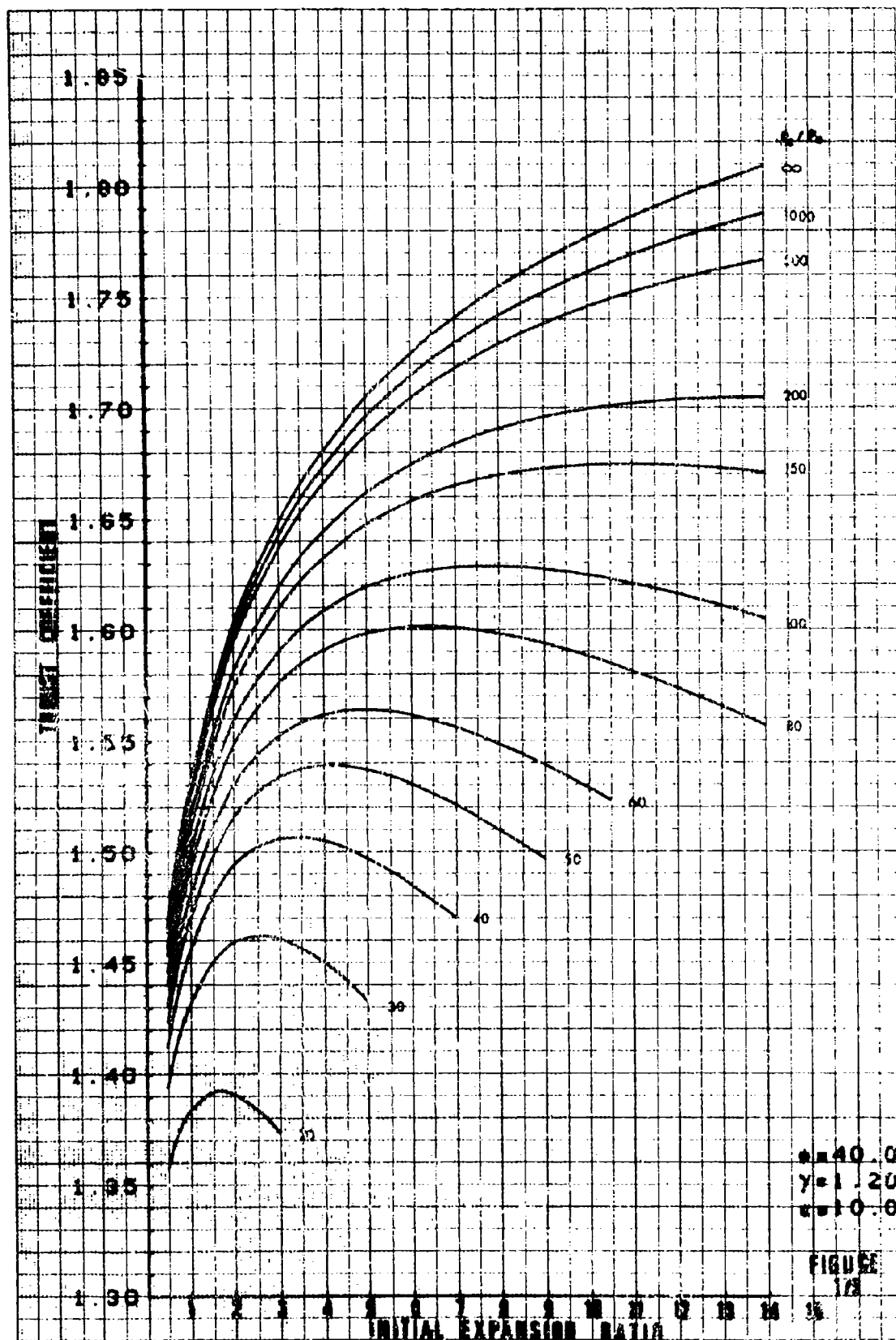


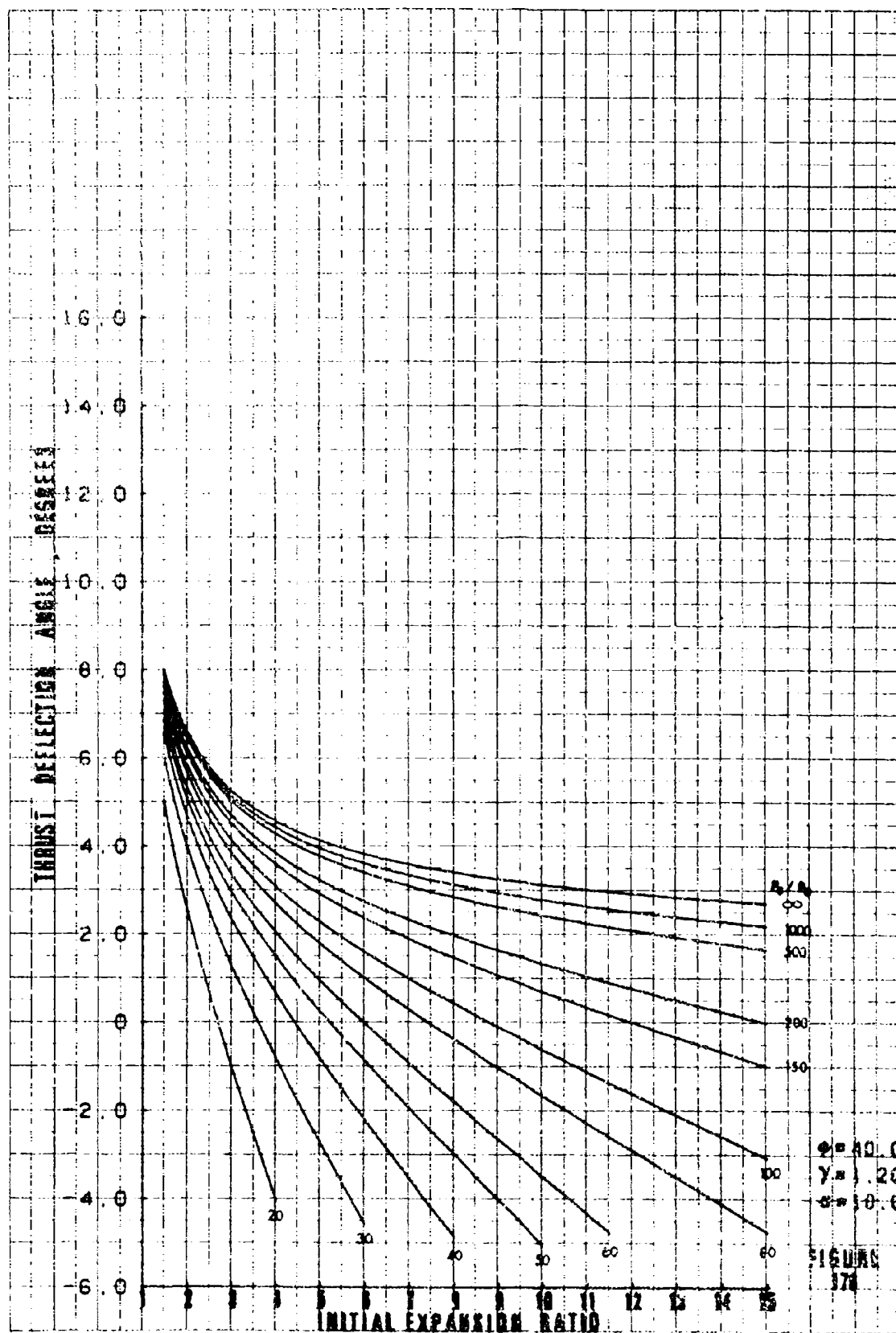
$\gamma = 1.20$
 $\sigma = 10.0$

FIGURE 172



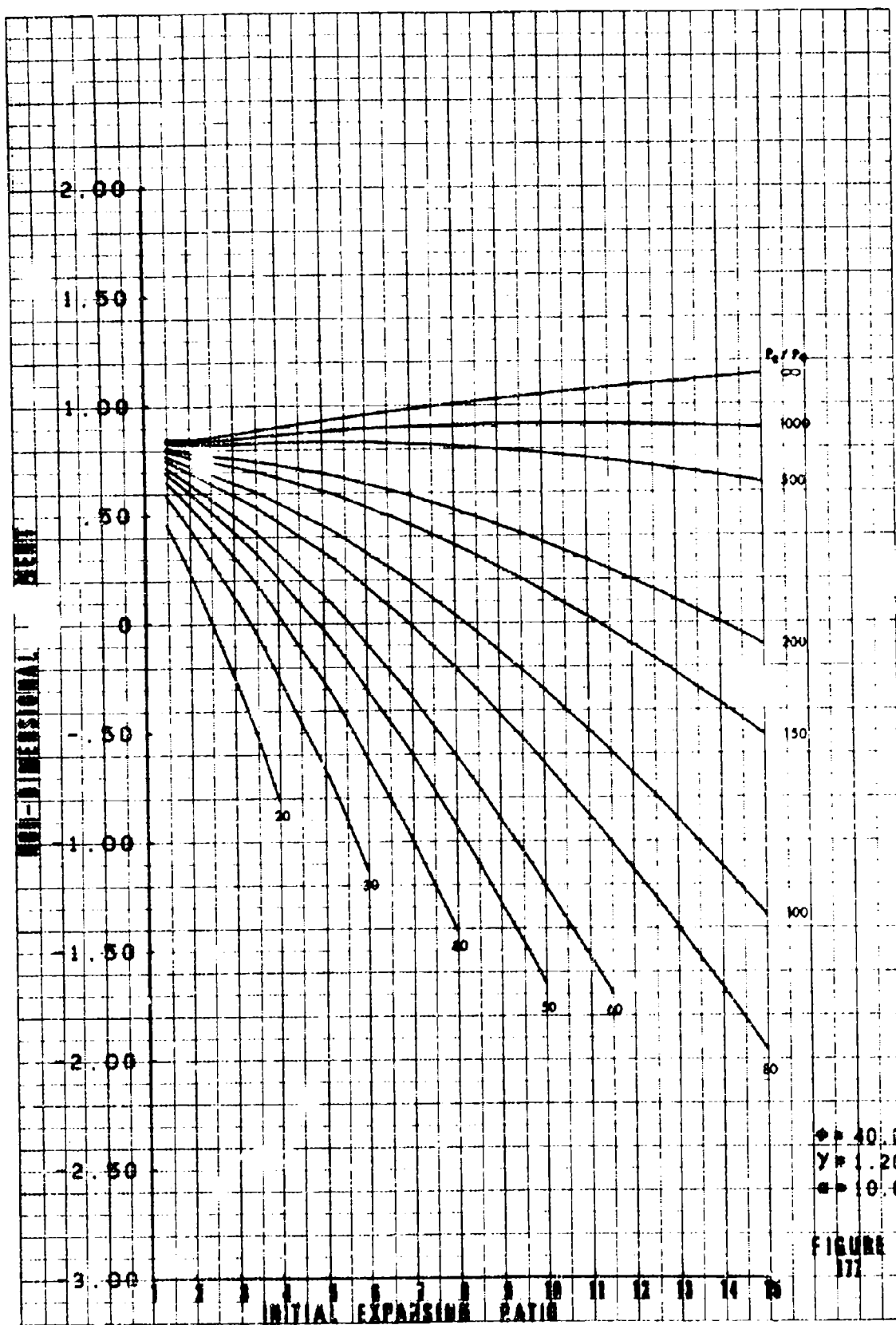


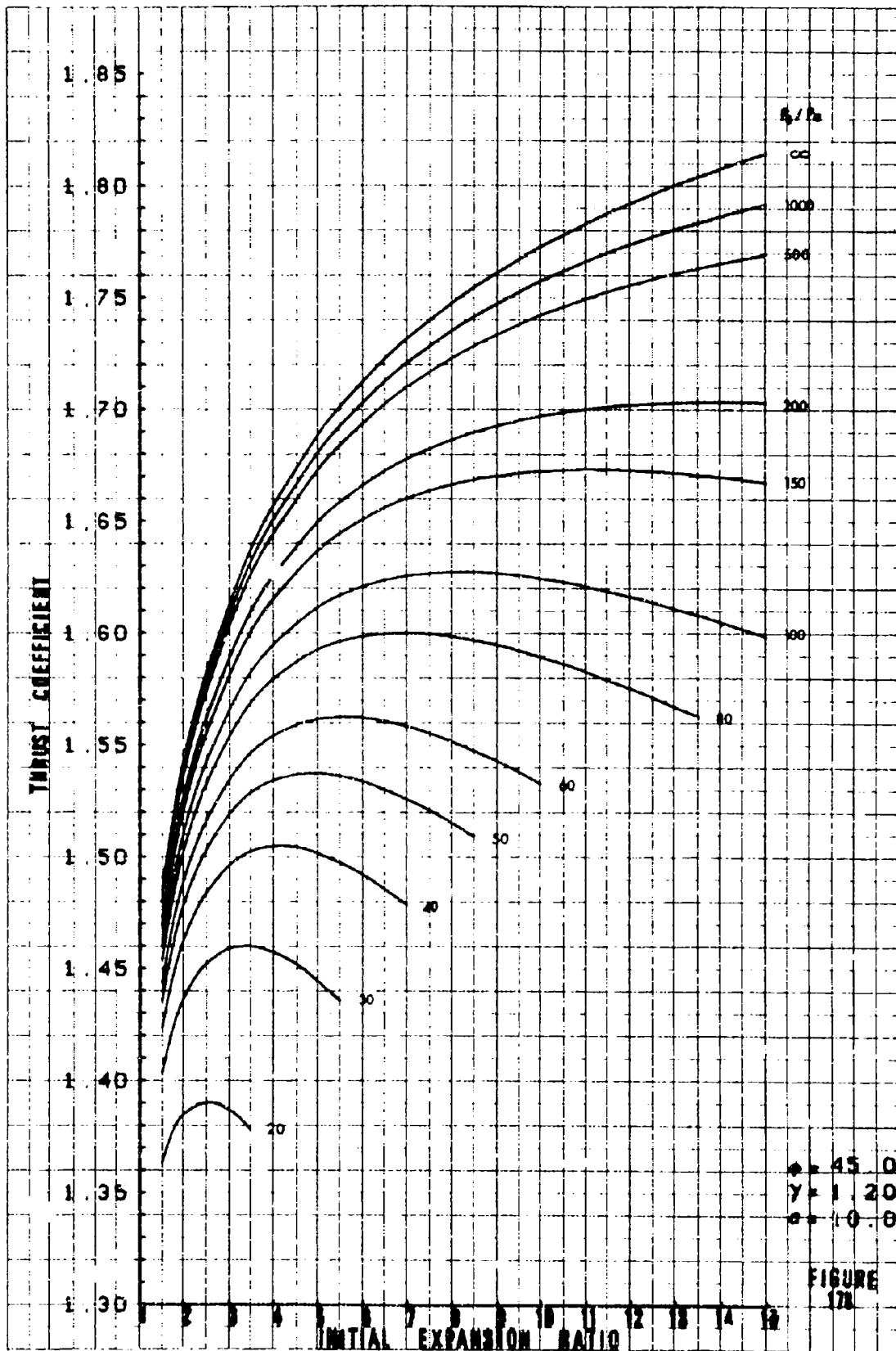




$\phi = 40.0$
 $\gamma = 1.20$
 $\sigma = 10.6$

FIGURE
 37





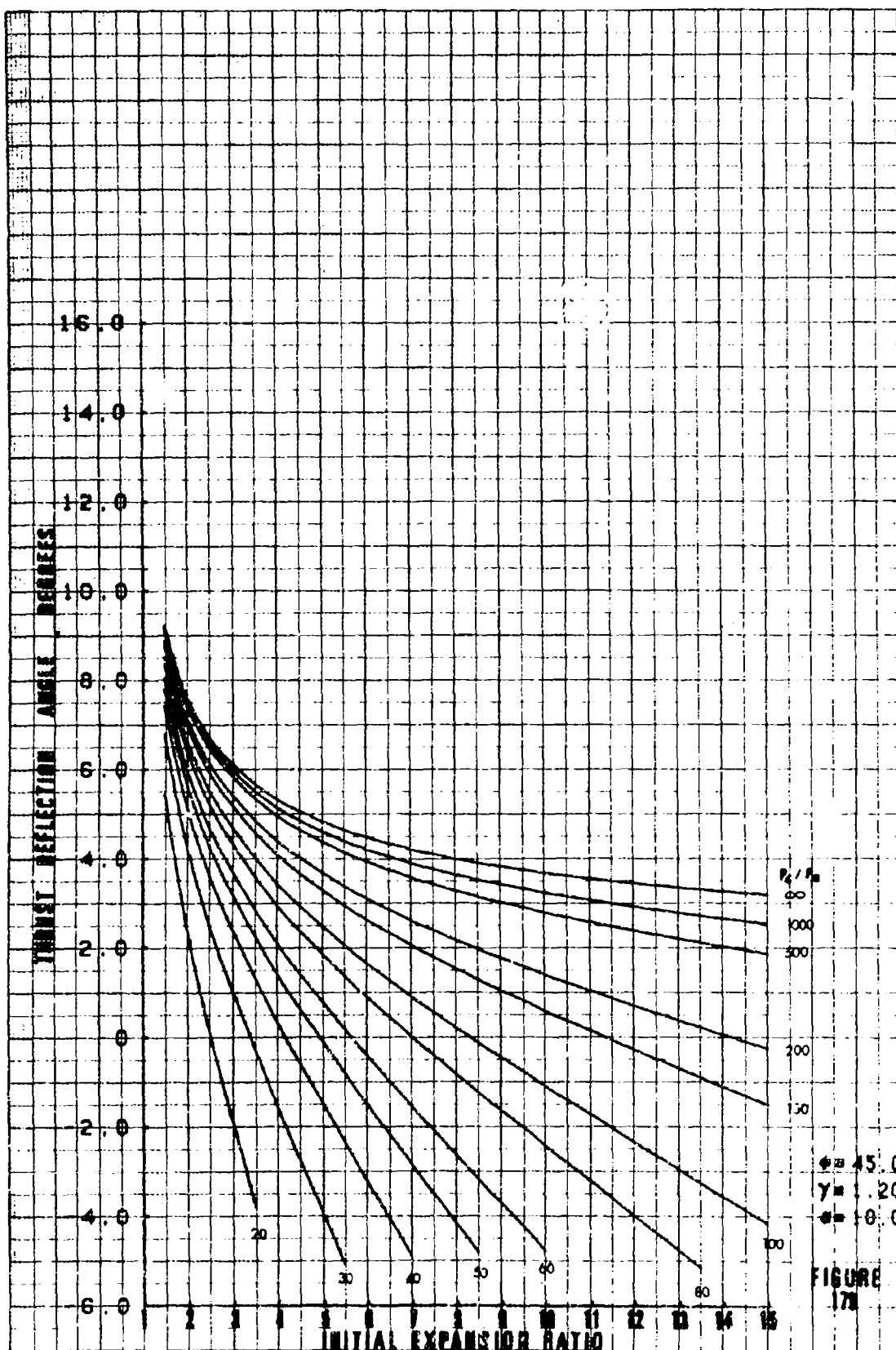
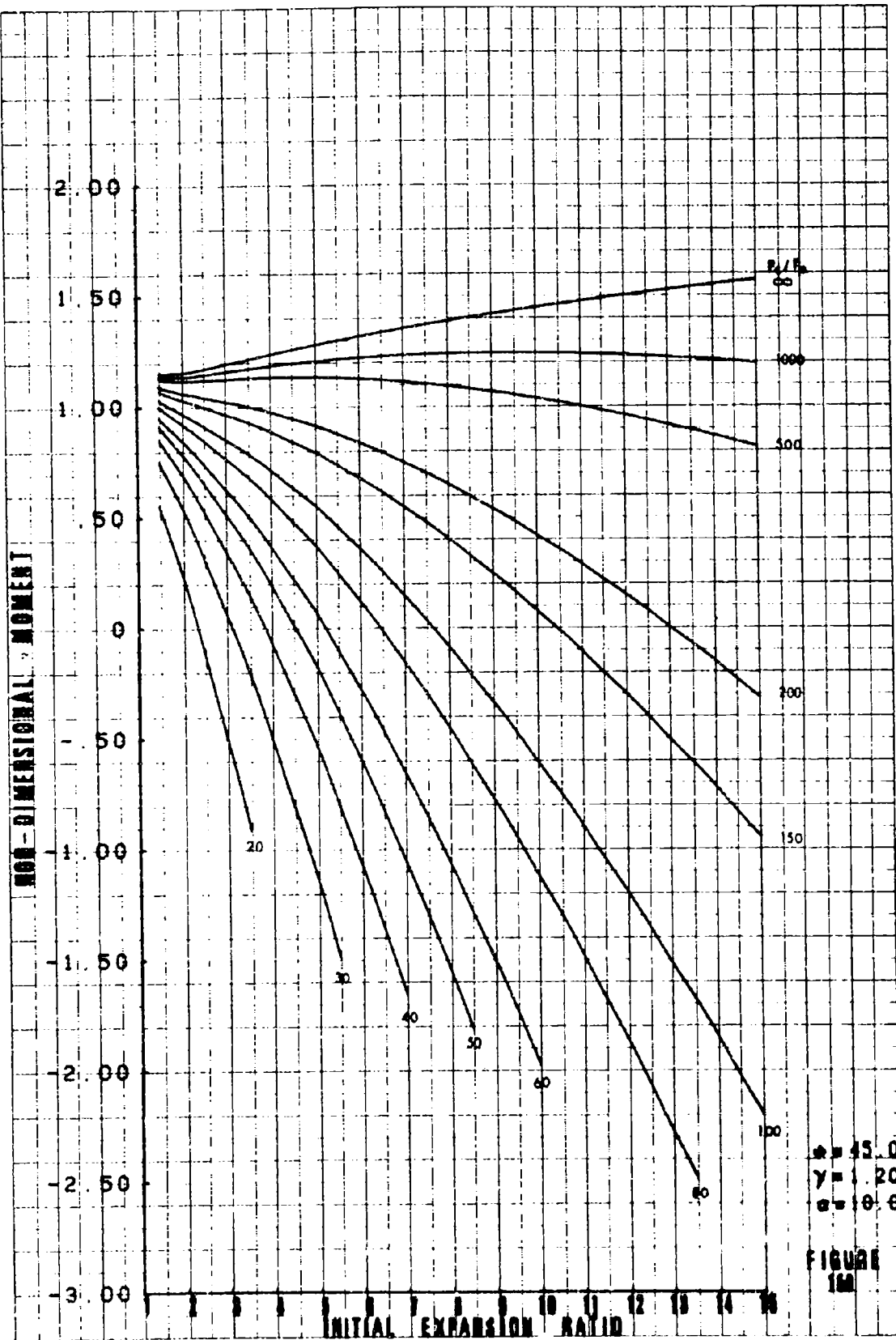
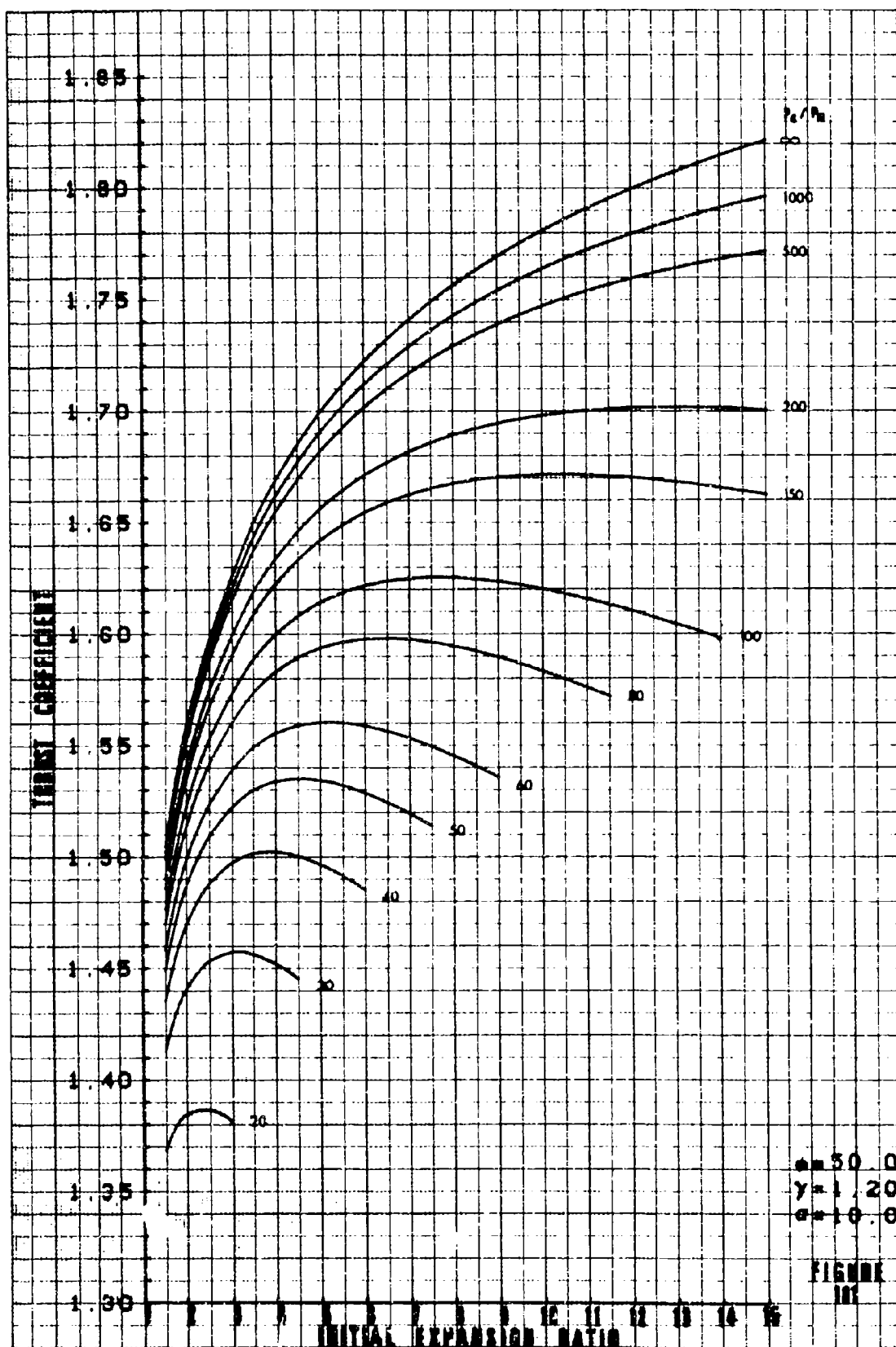
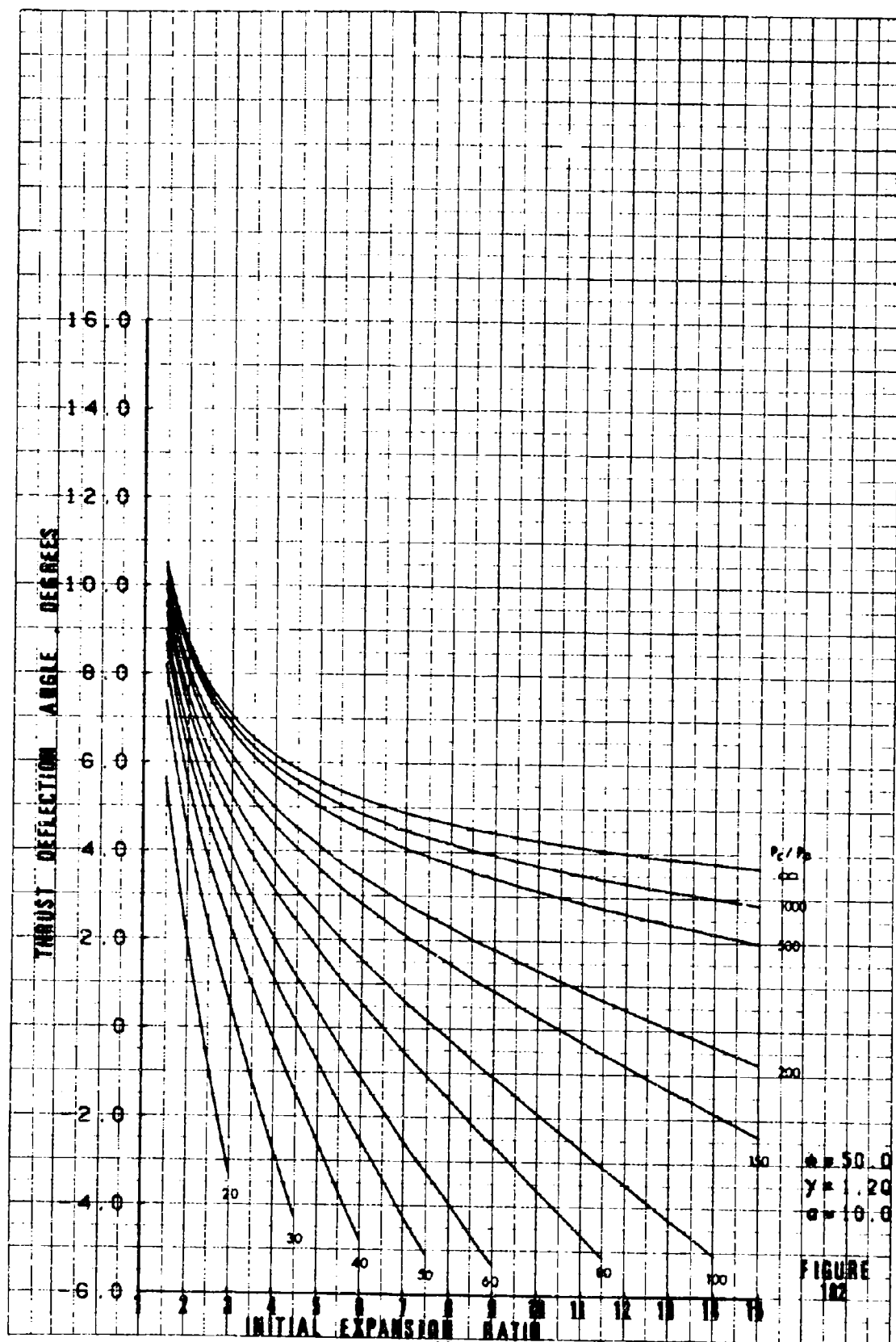
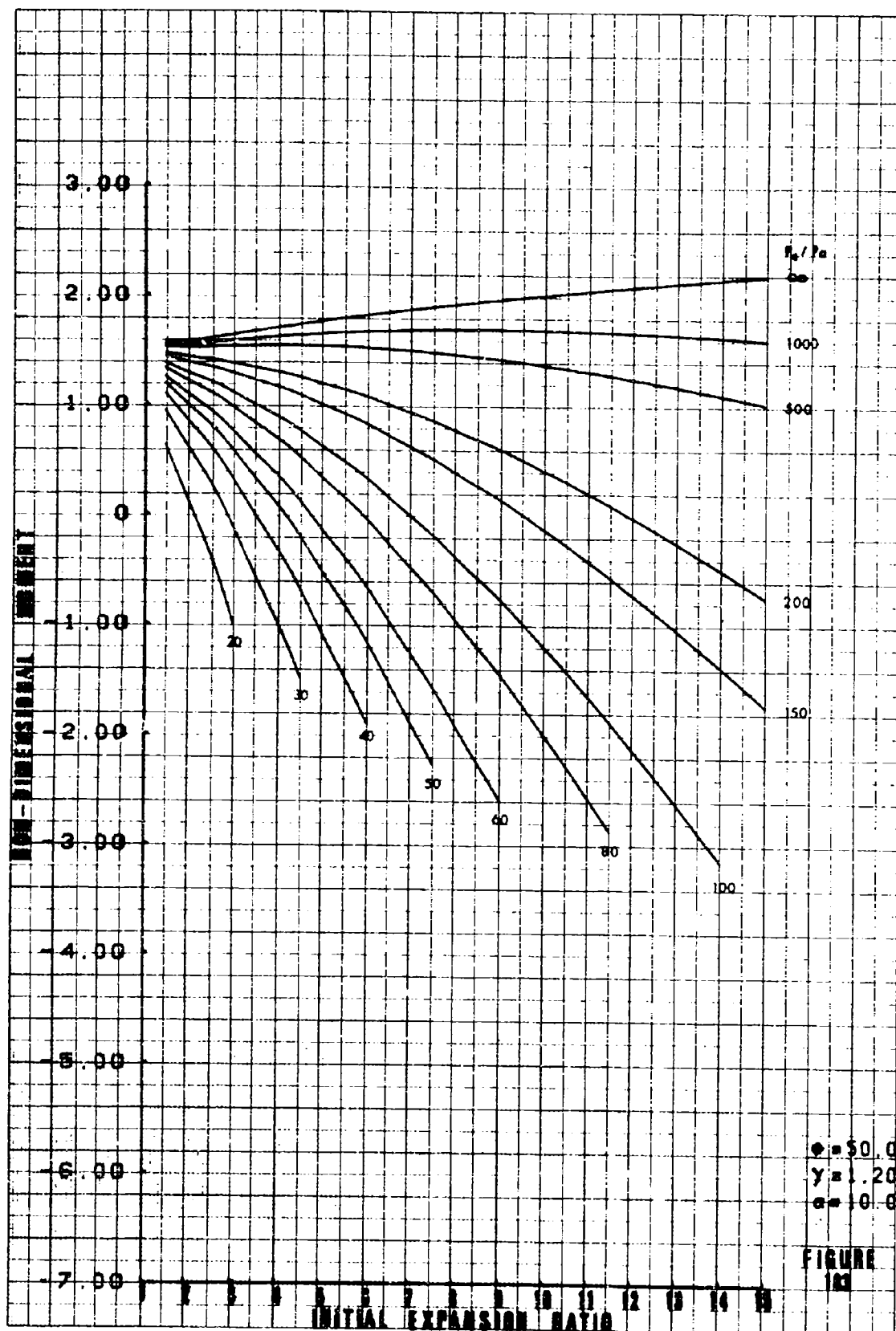


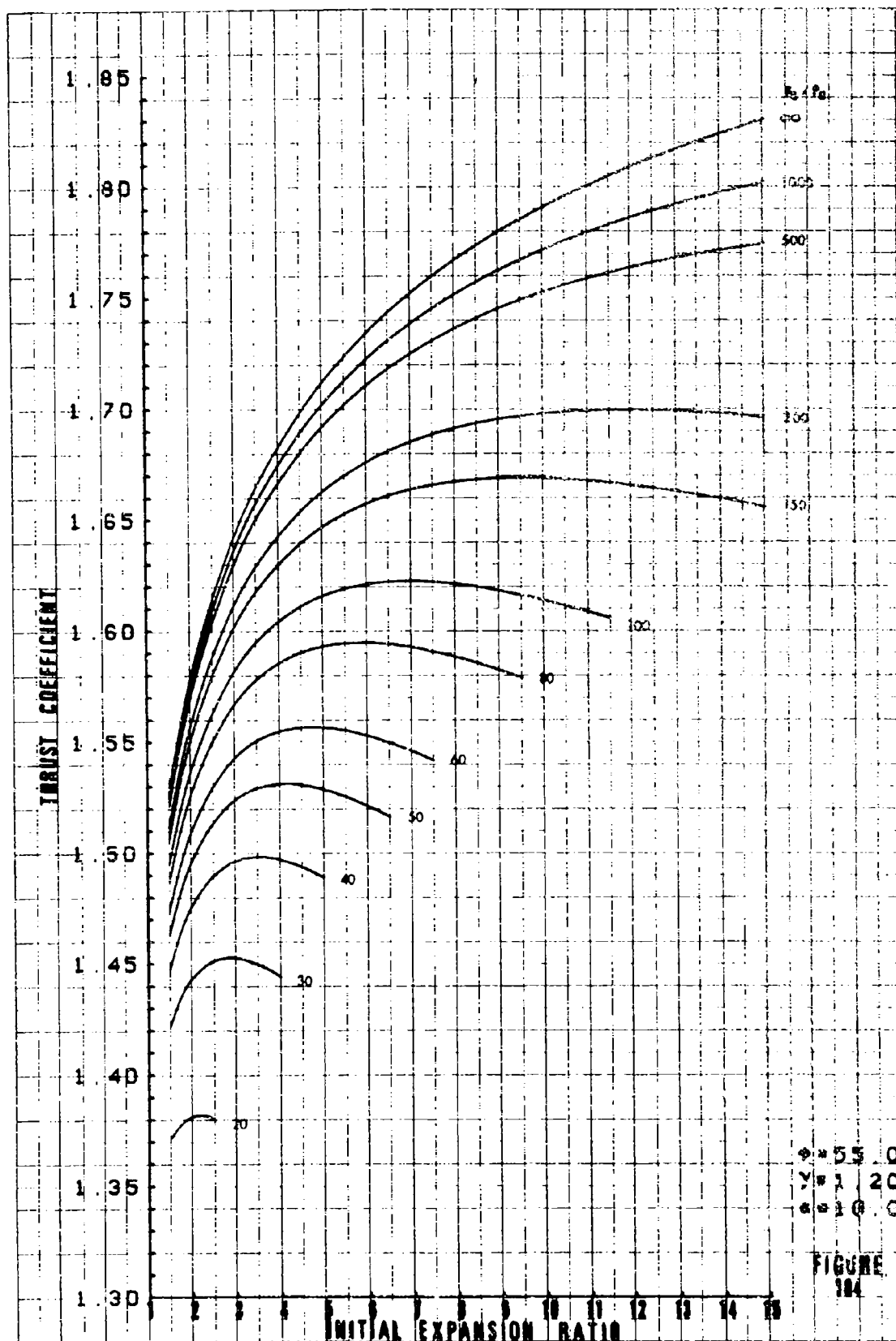
FIGURE 17





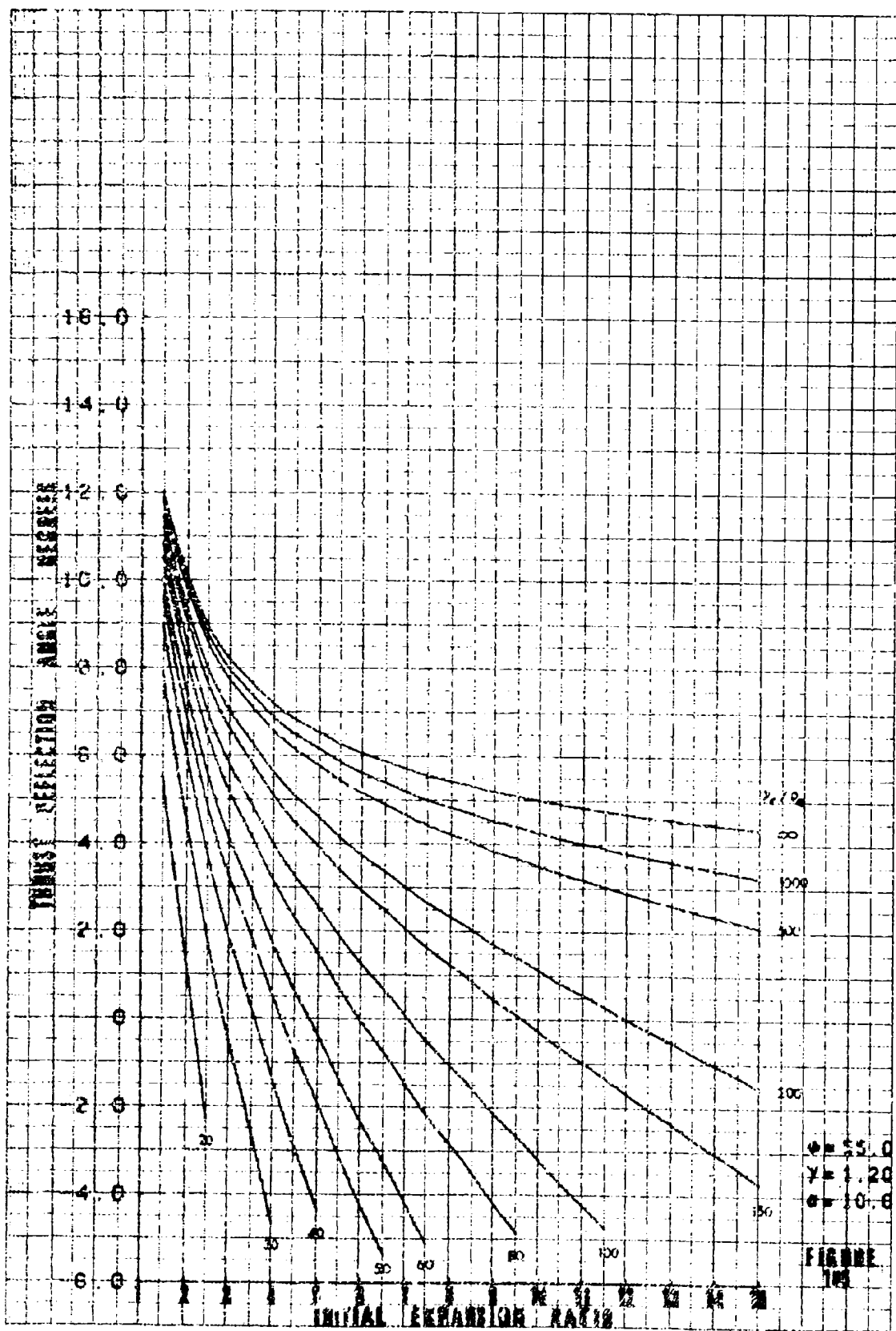


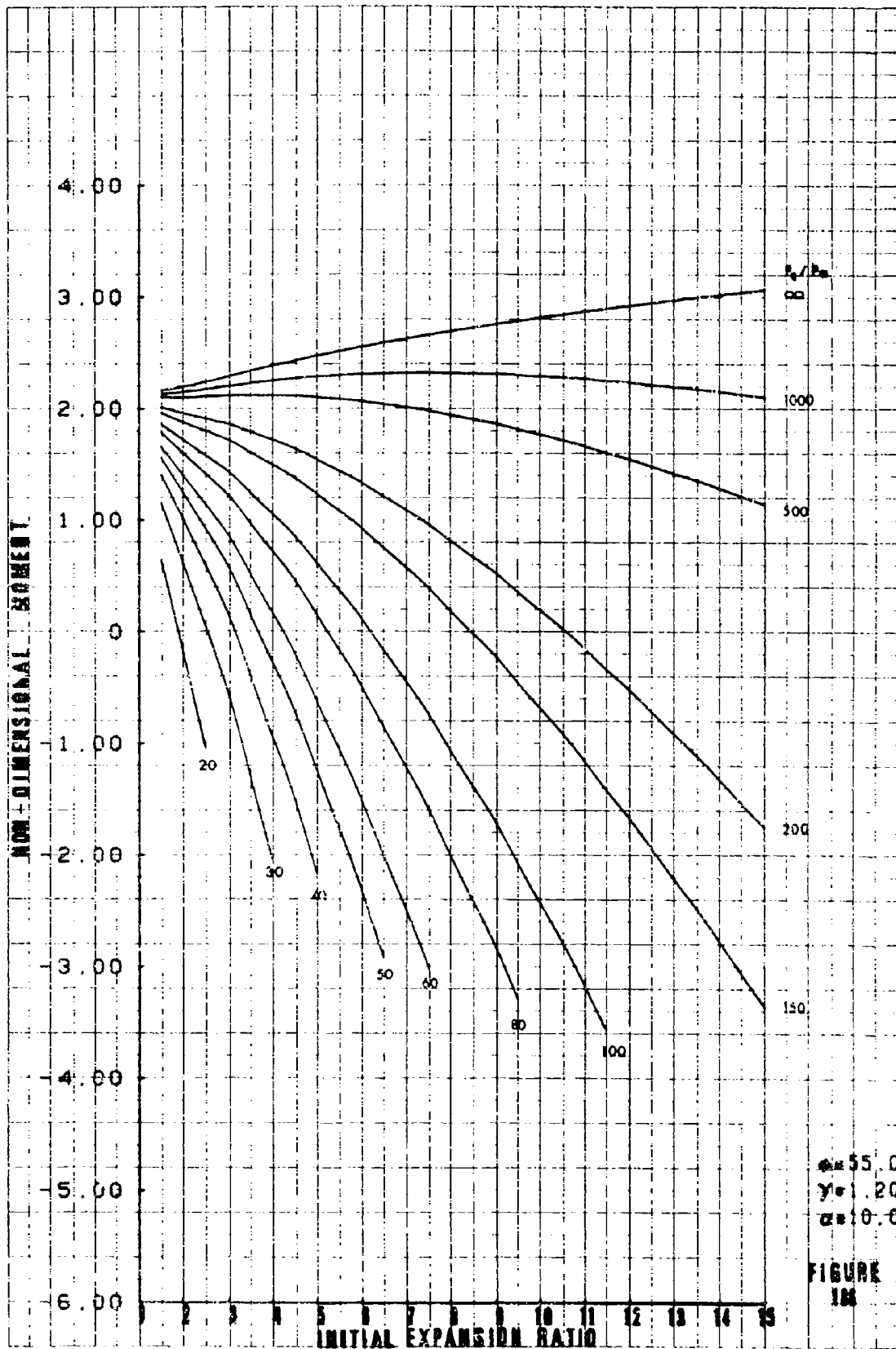




$\phi = 0.55$
 $\gamma = 1.20$
 $\epsilon = 1.0$

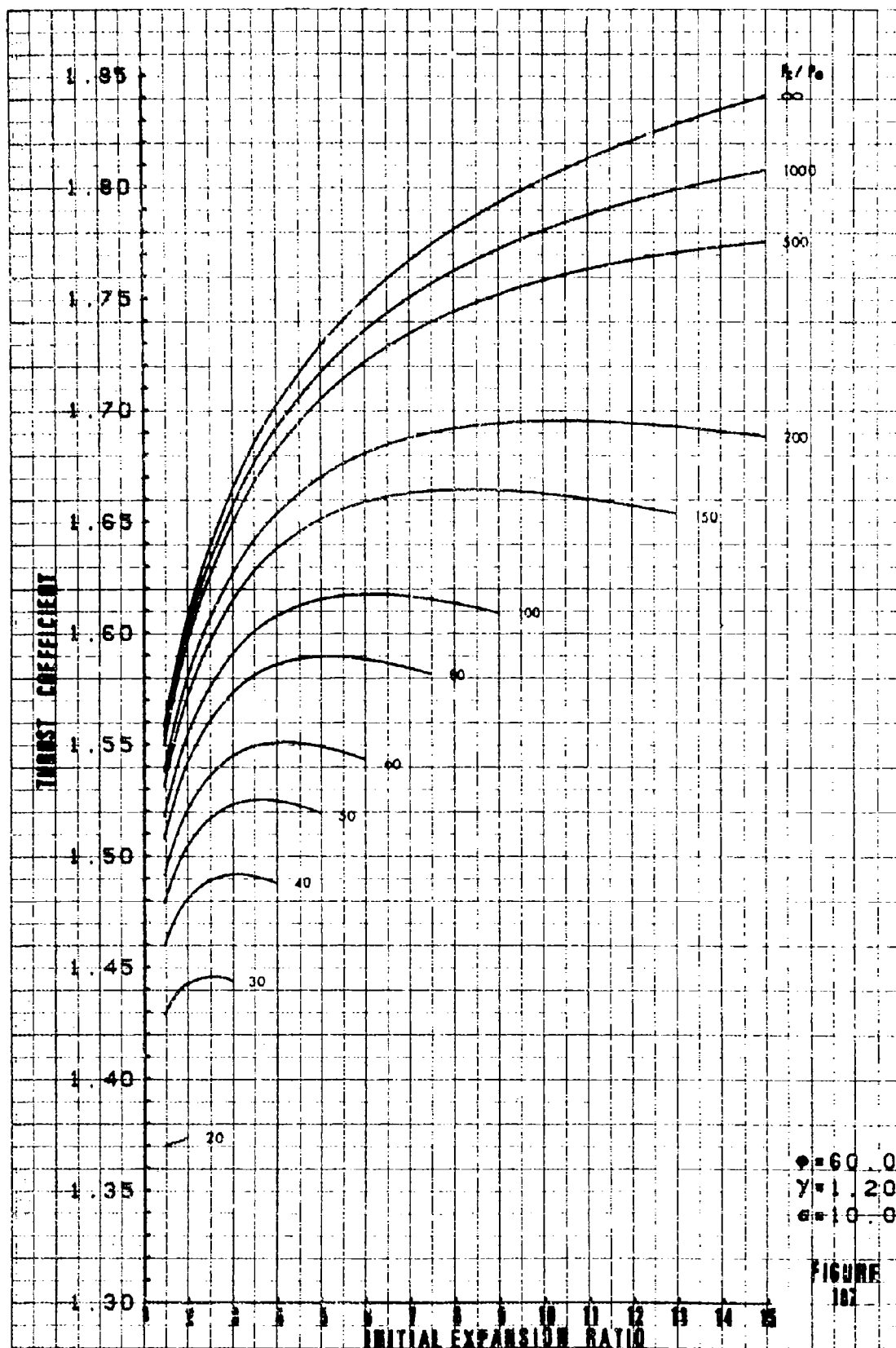
FIGURE 184

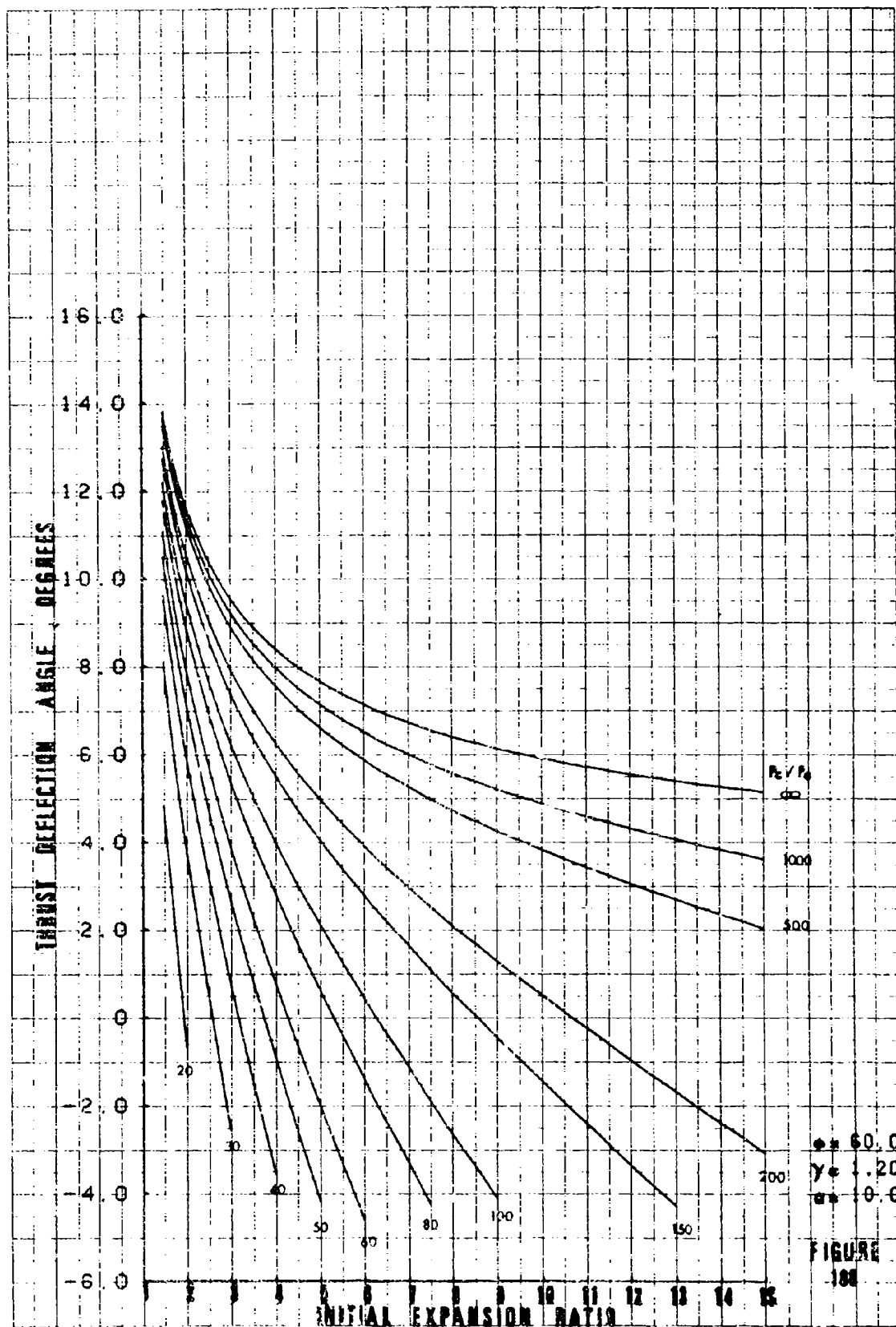


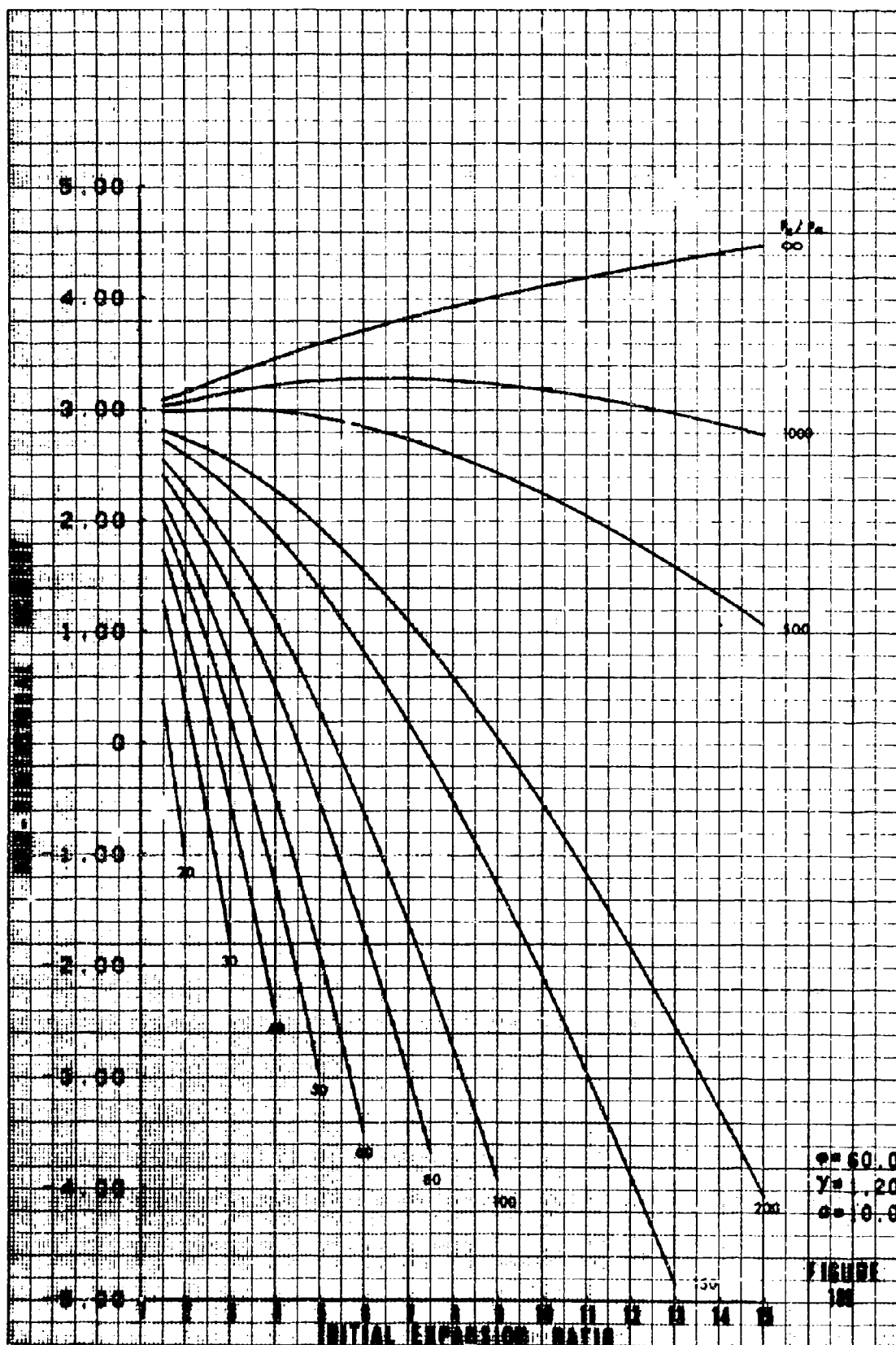


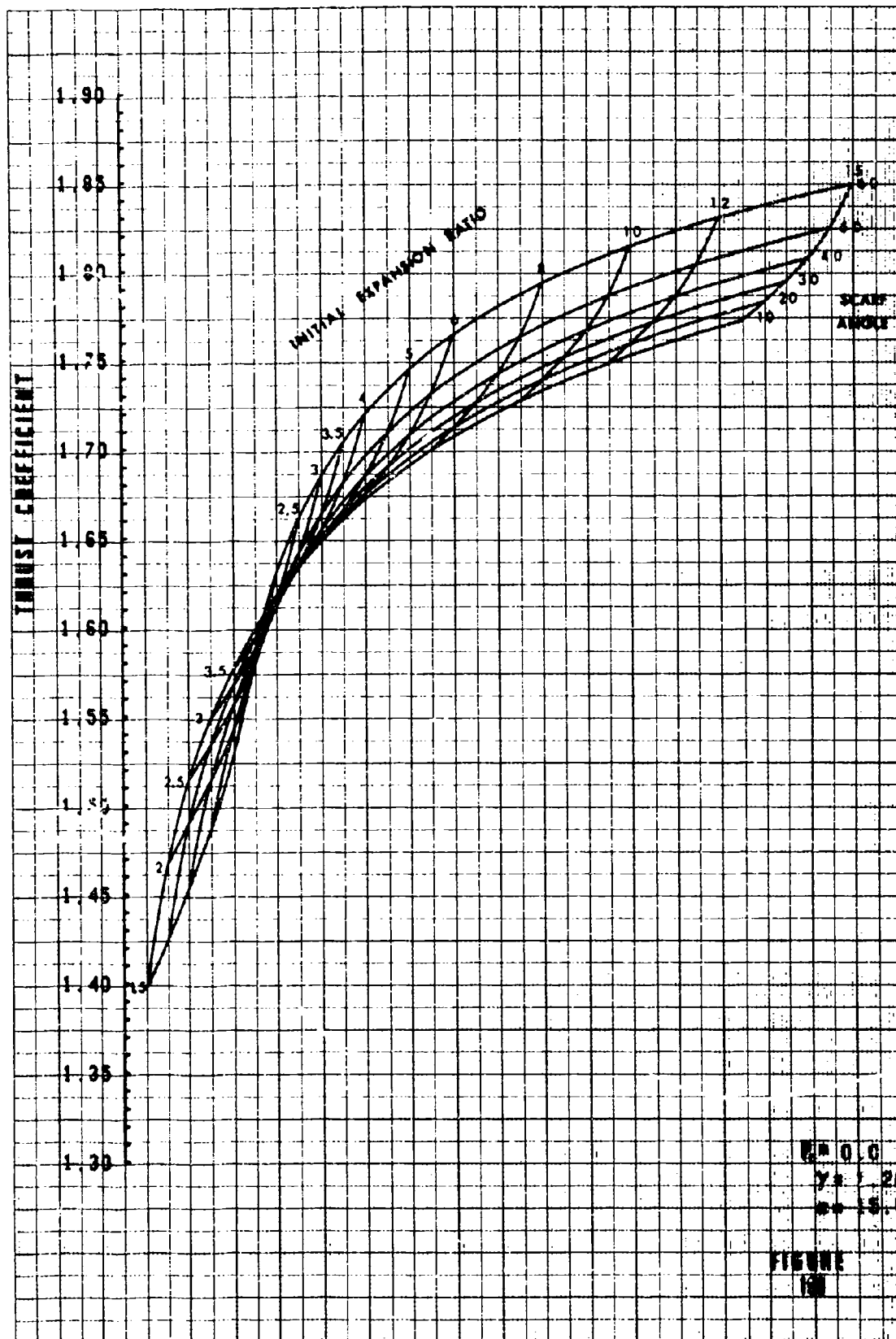
$\mu = 55.0$
 $\gamma = 1.20$
 $\alpha = 0.0$

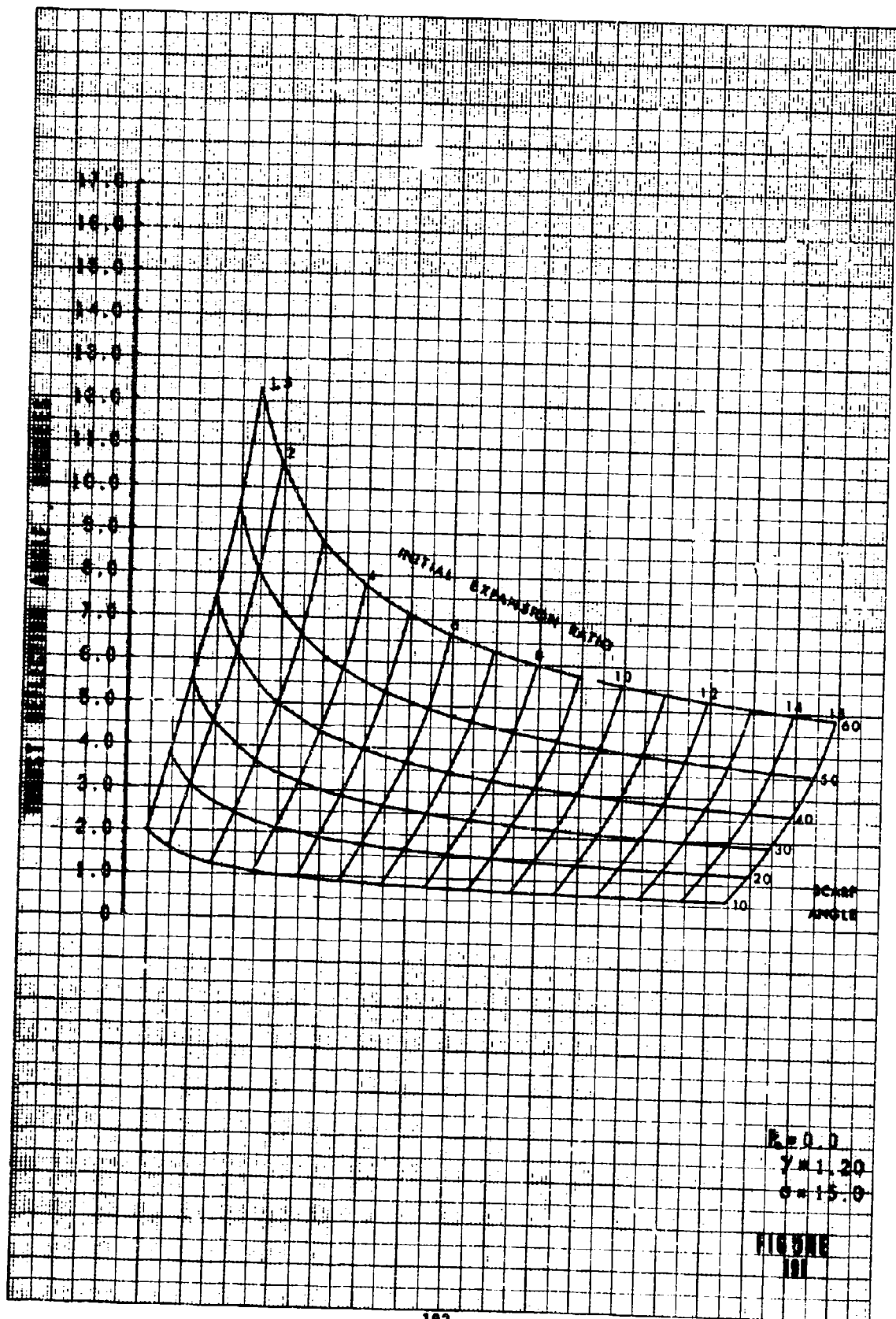
FIGURE 10A

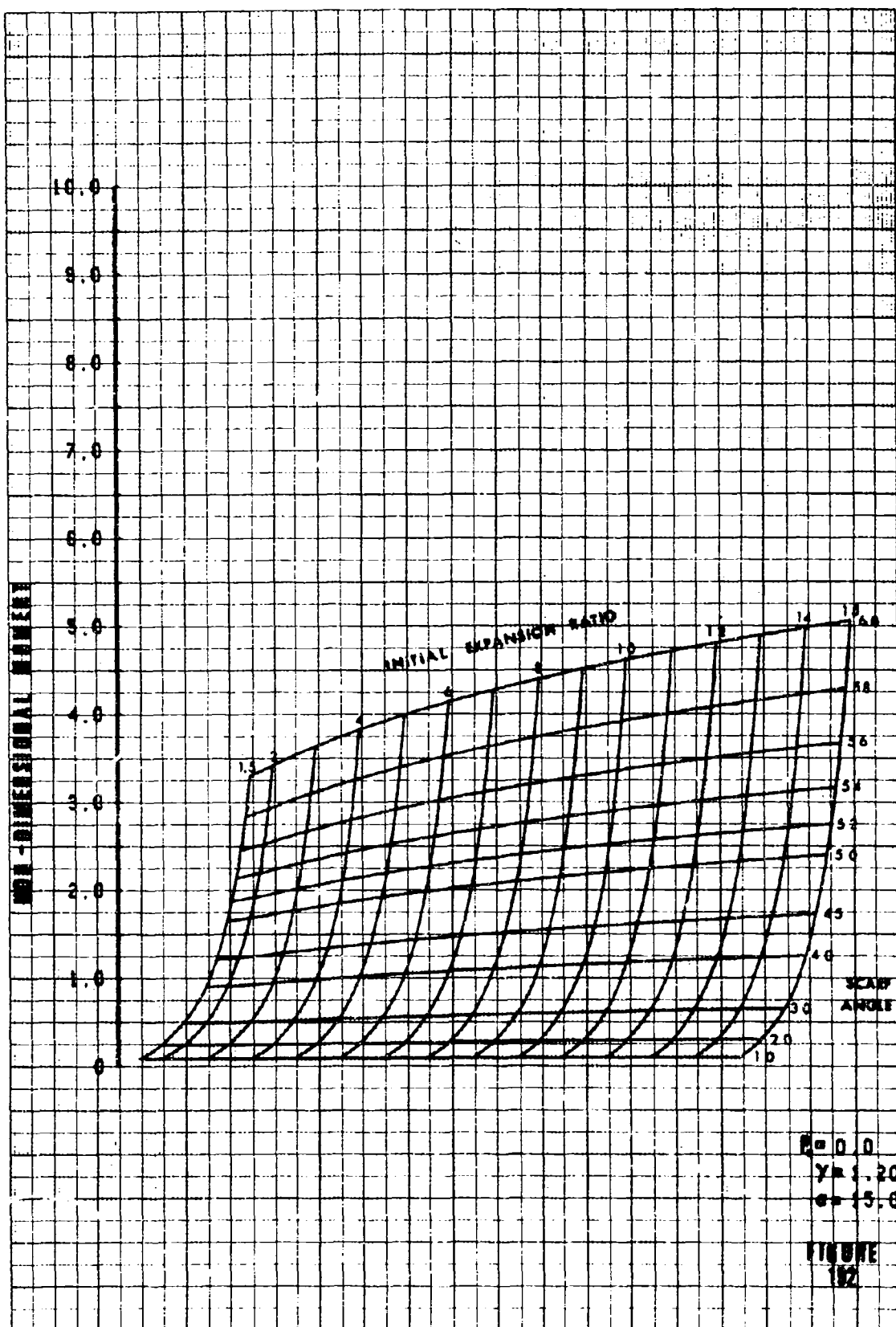


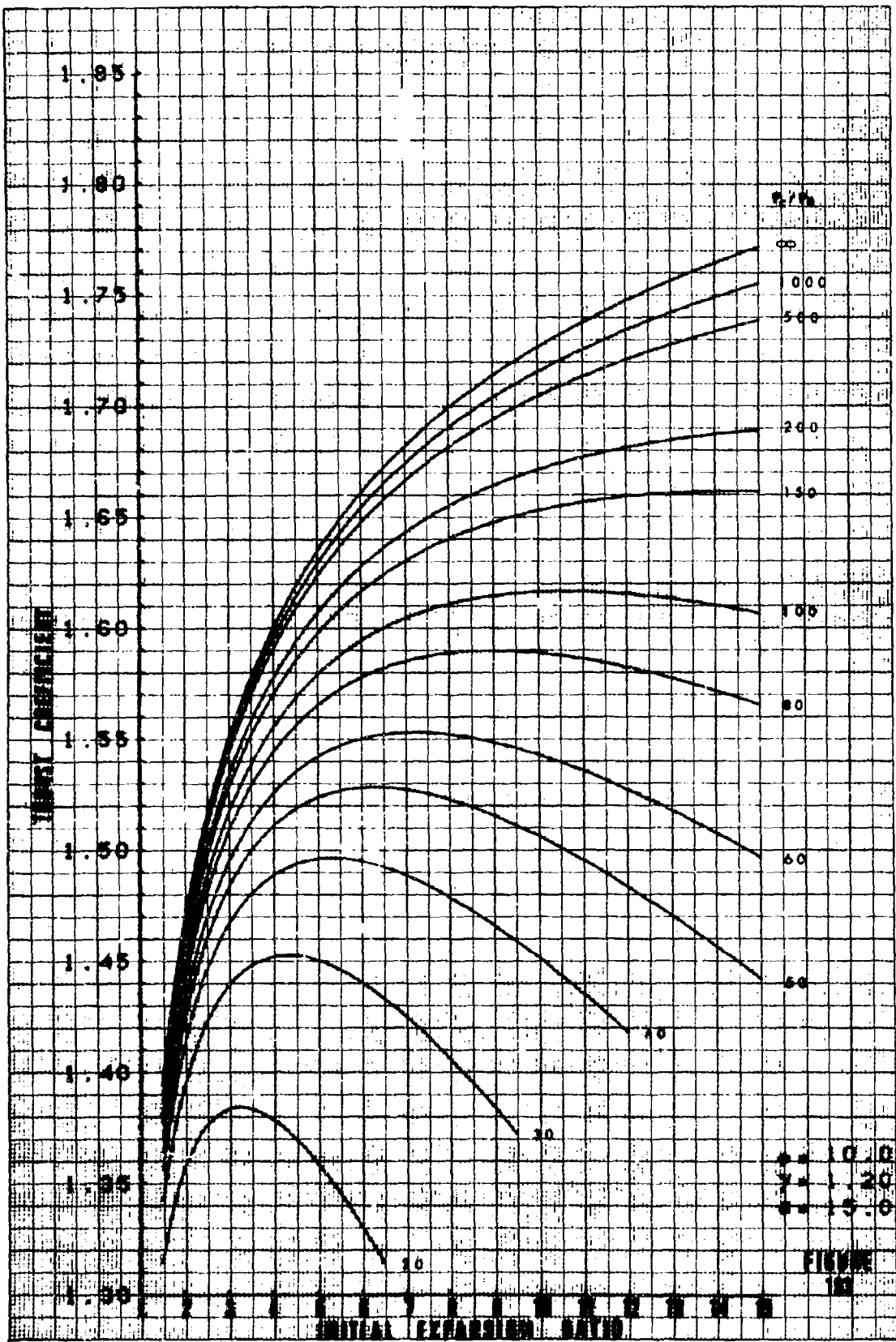


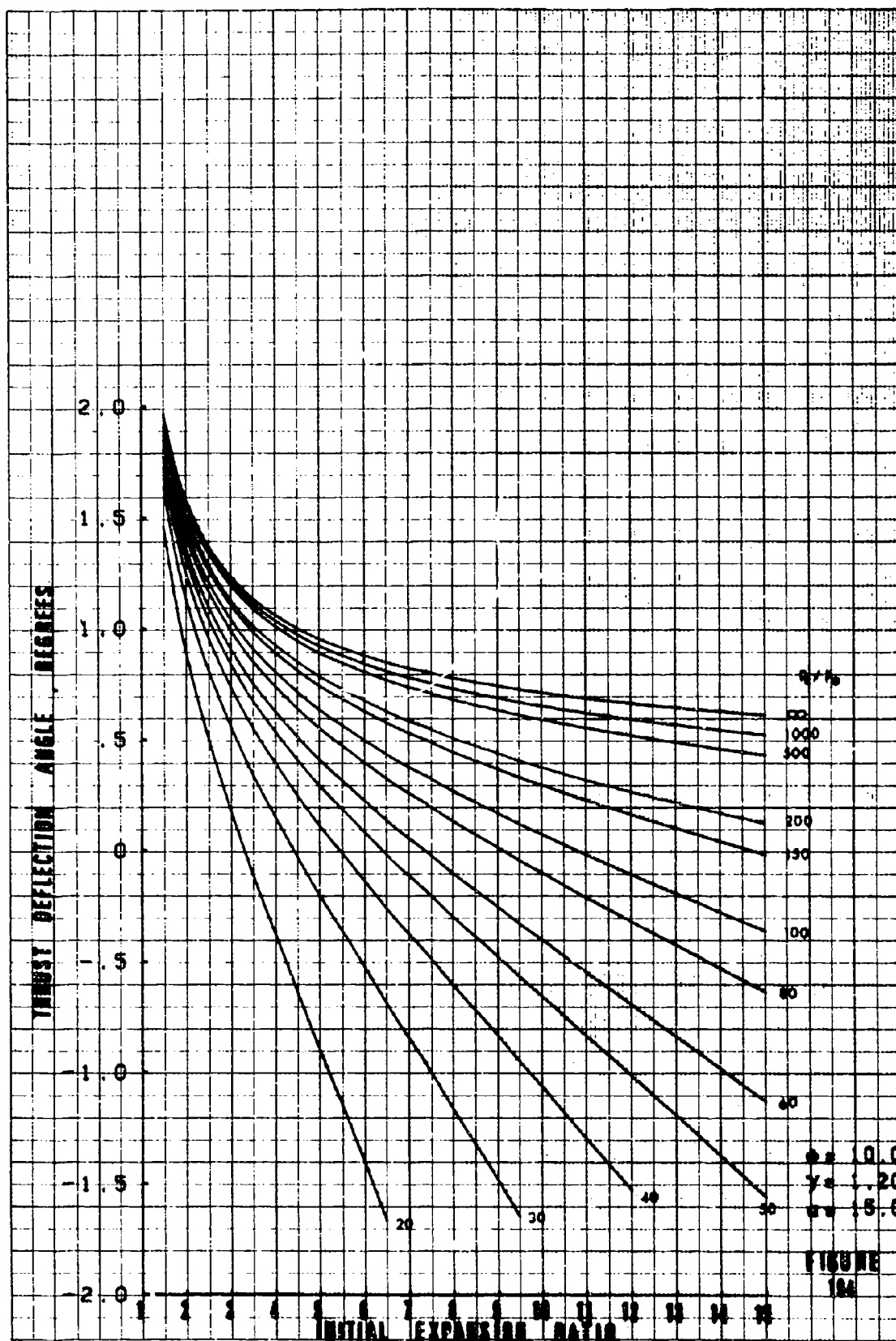












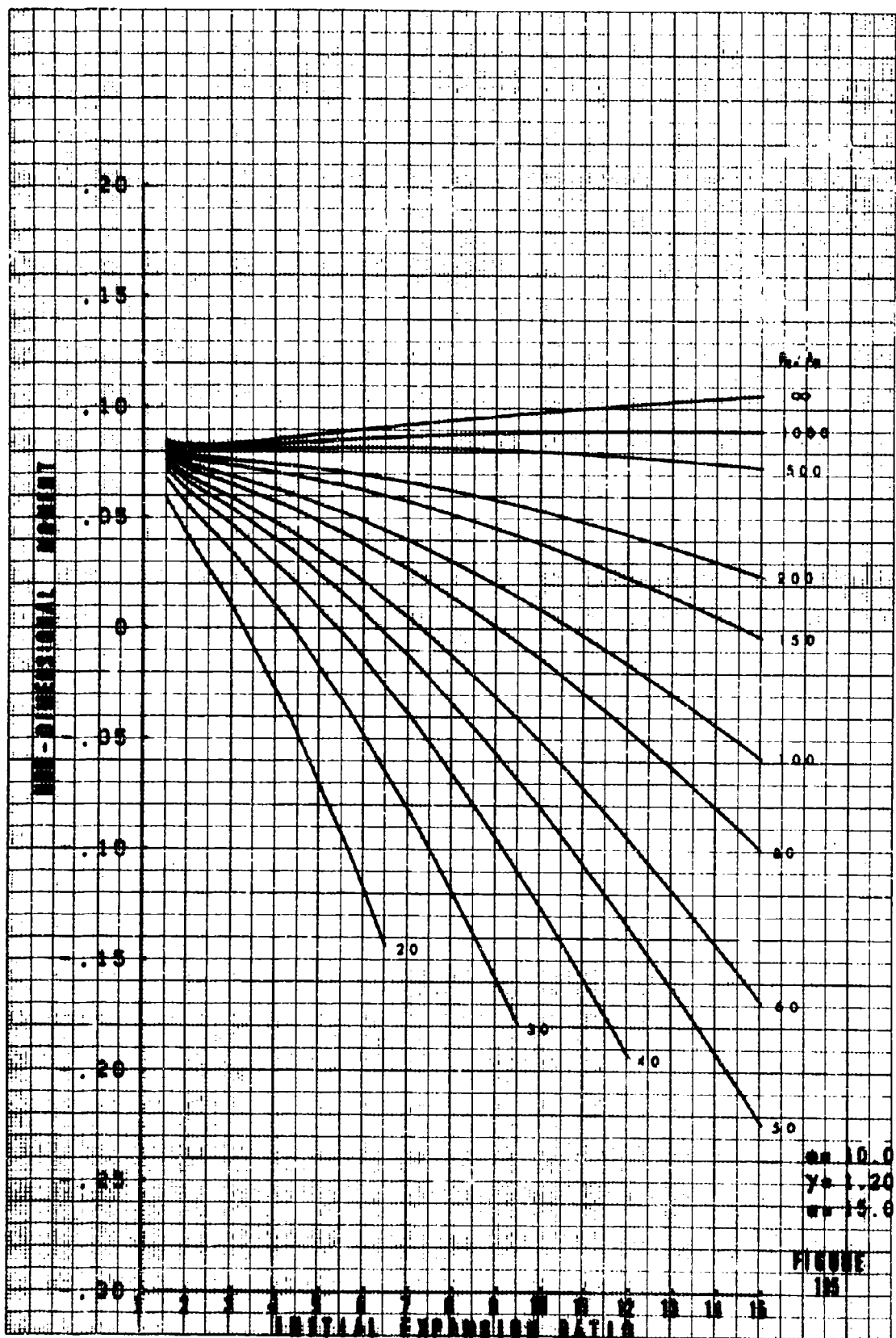
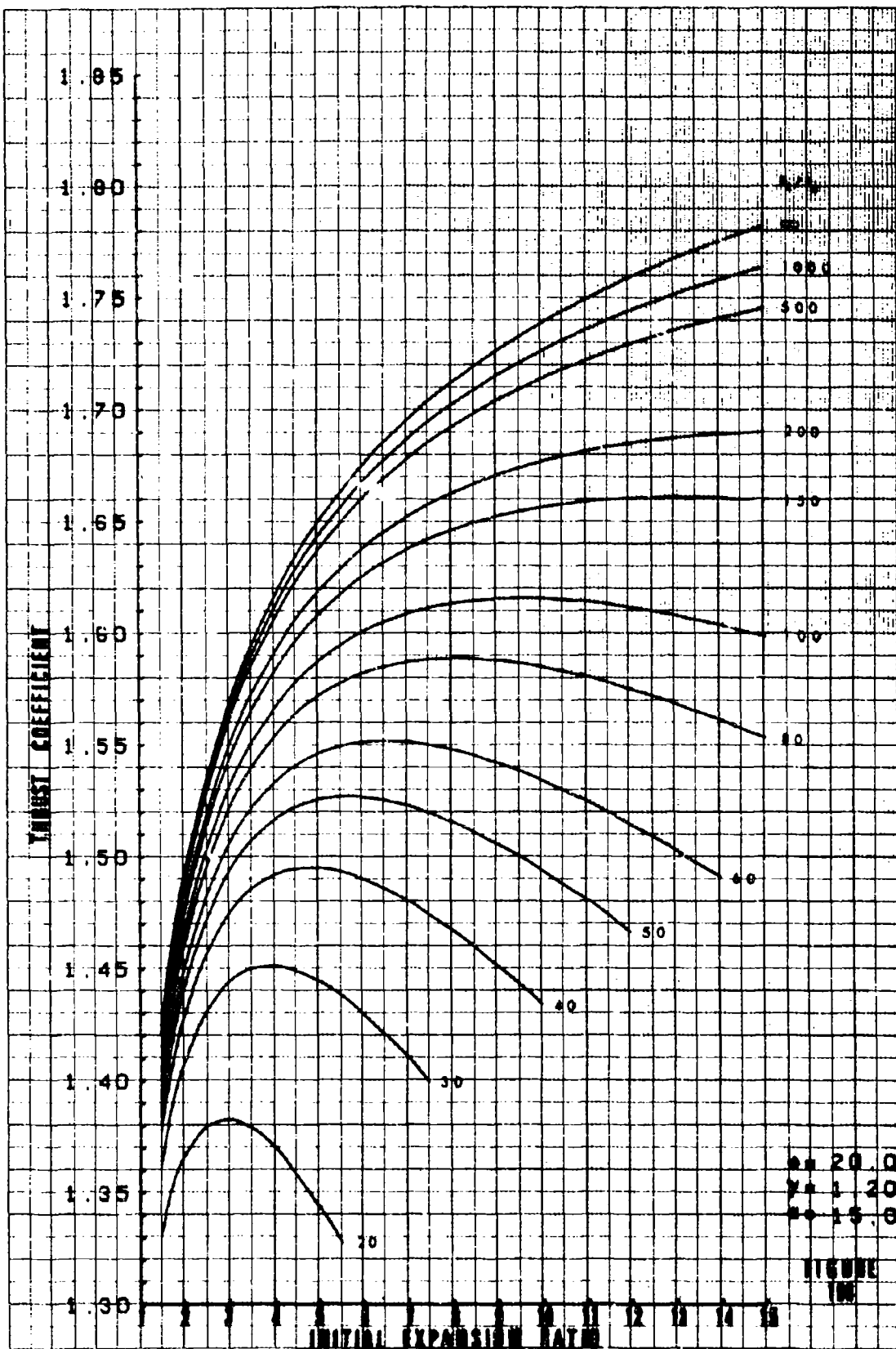
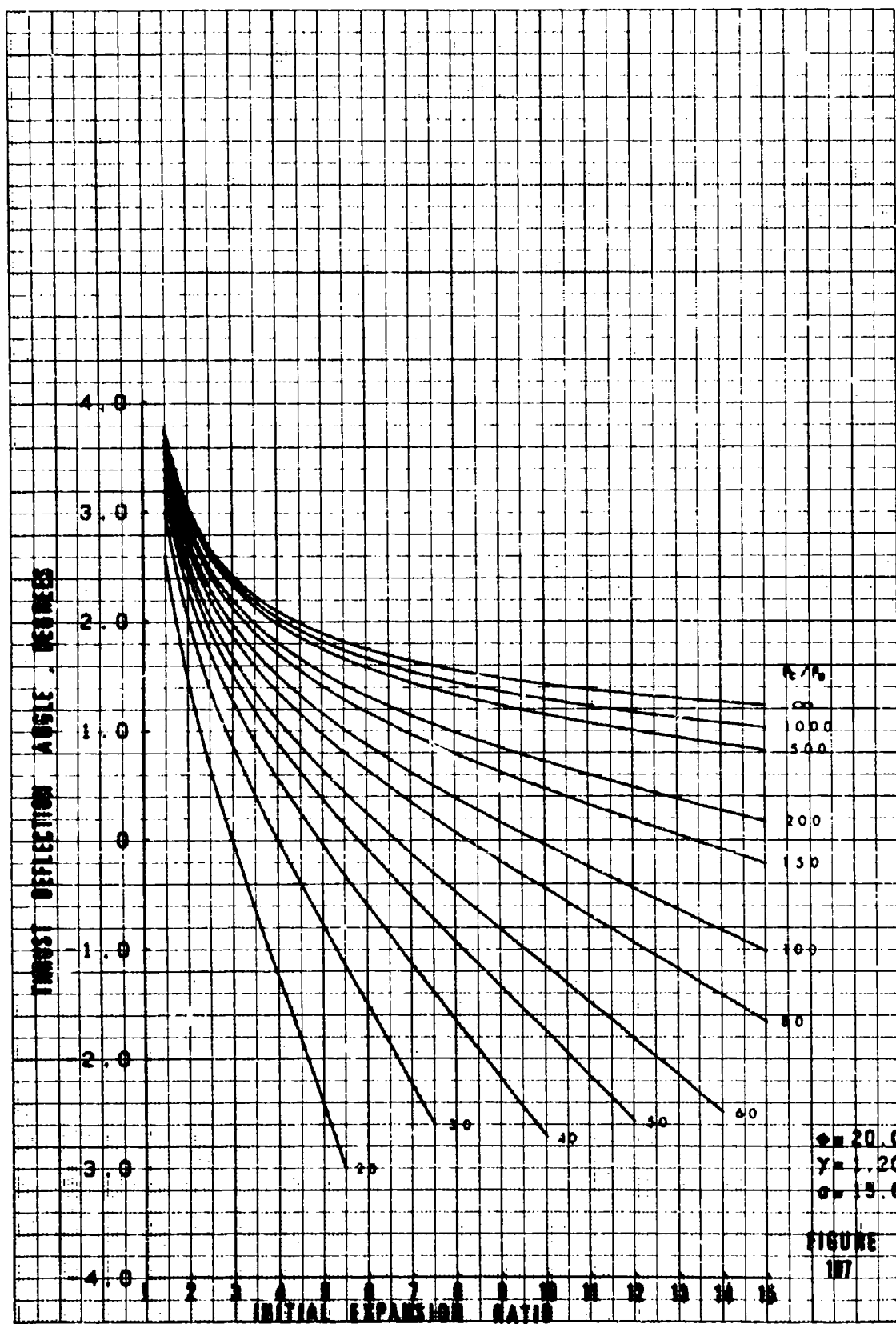


FIGURE 15





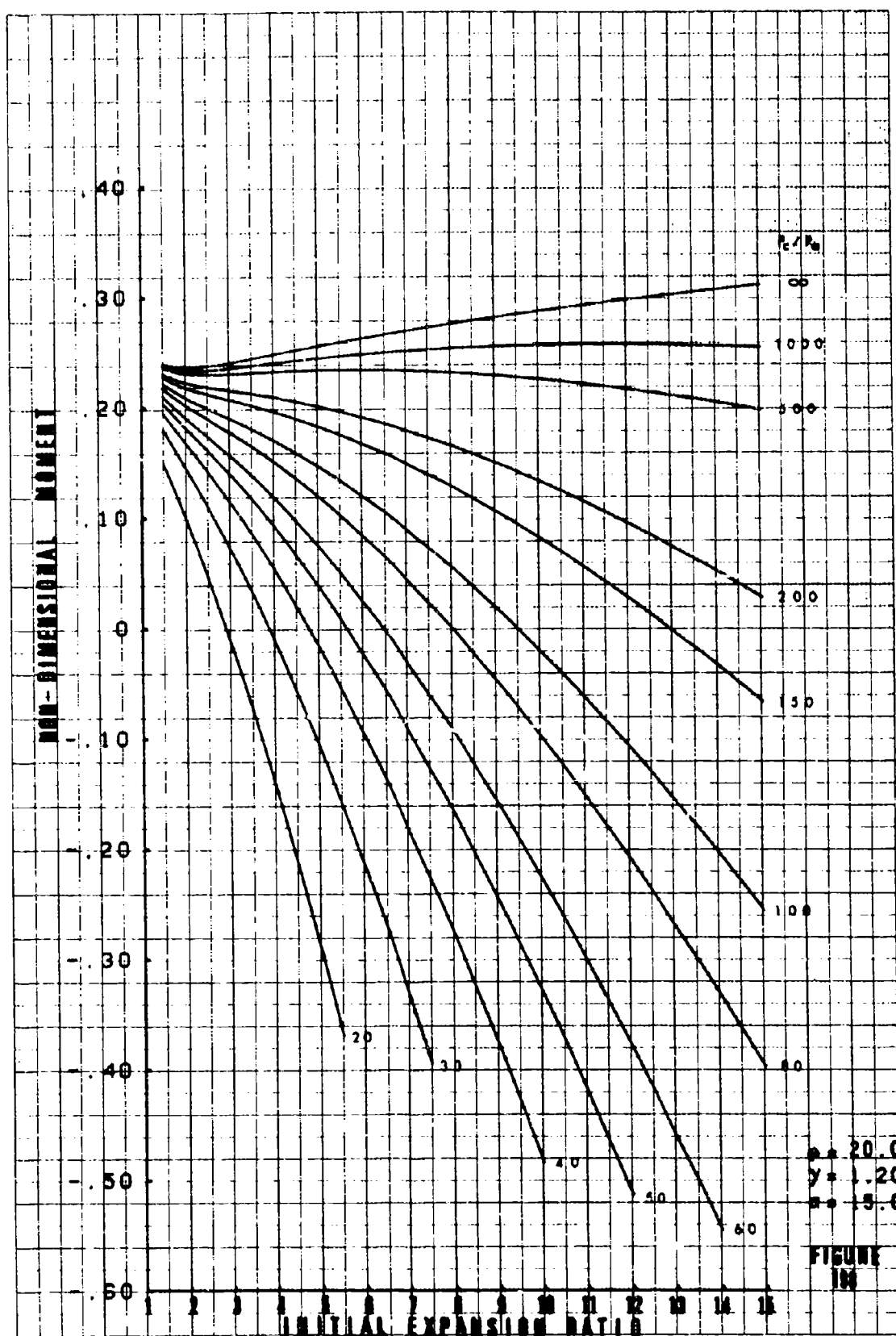
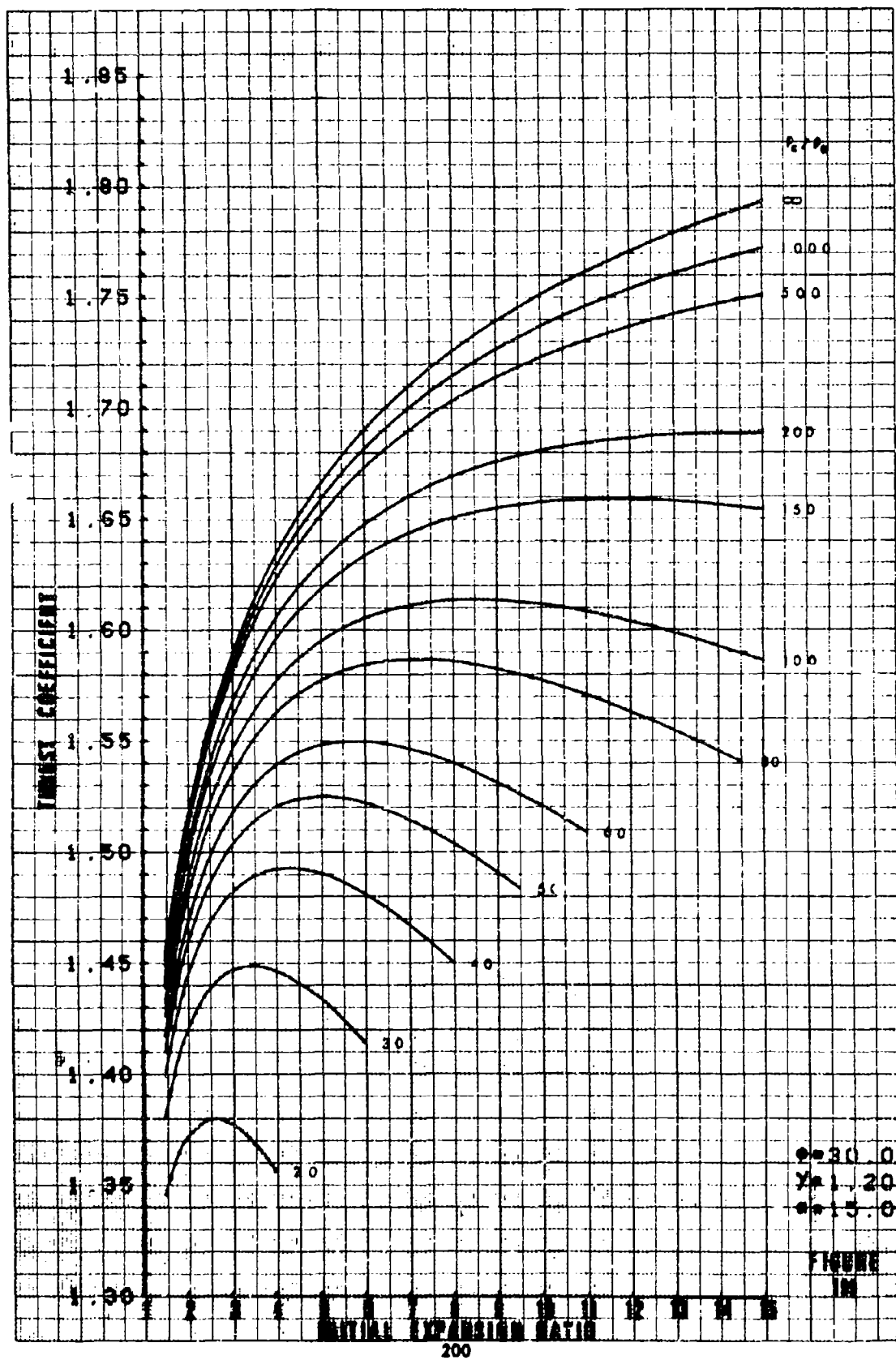
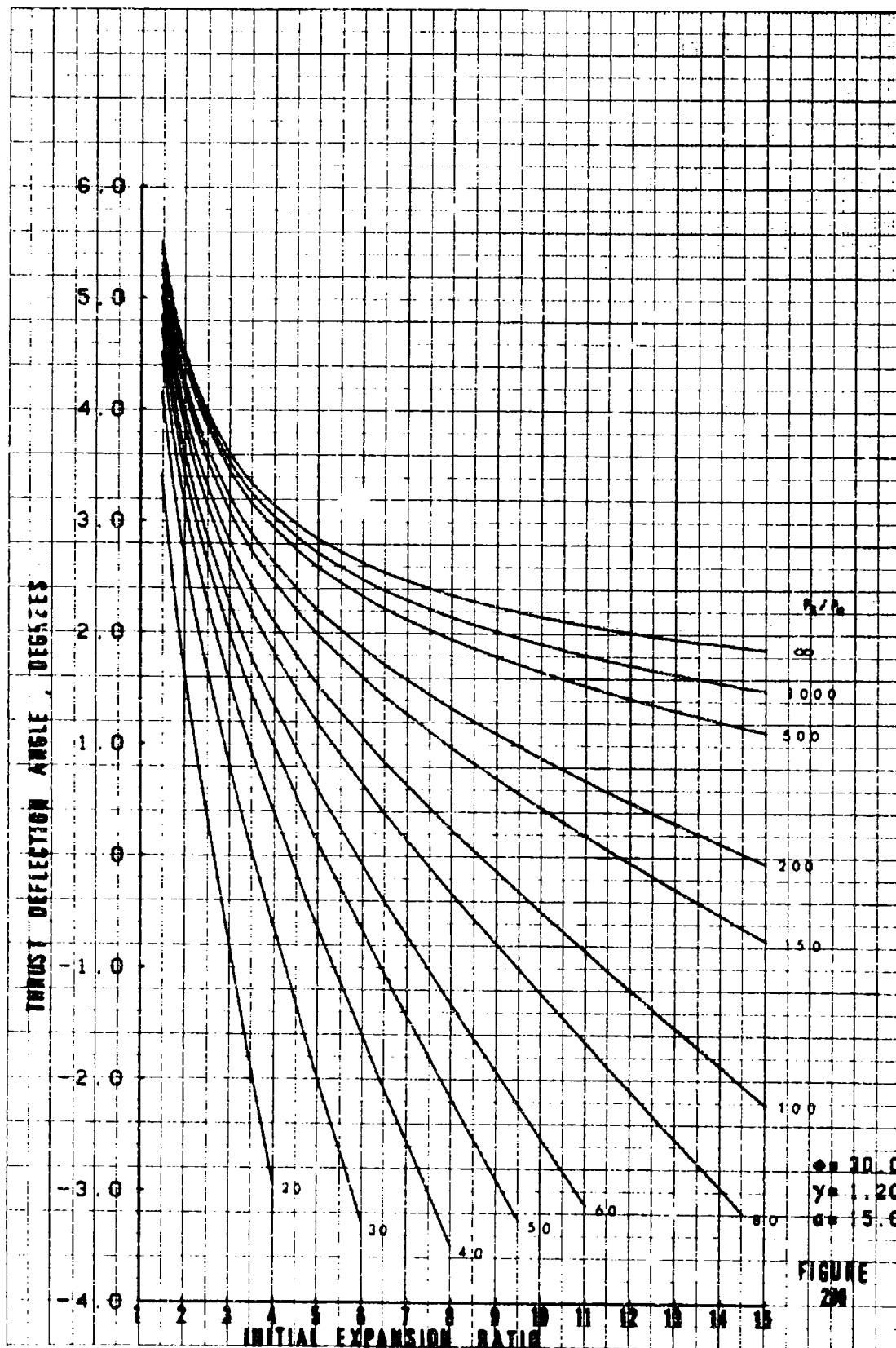
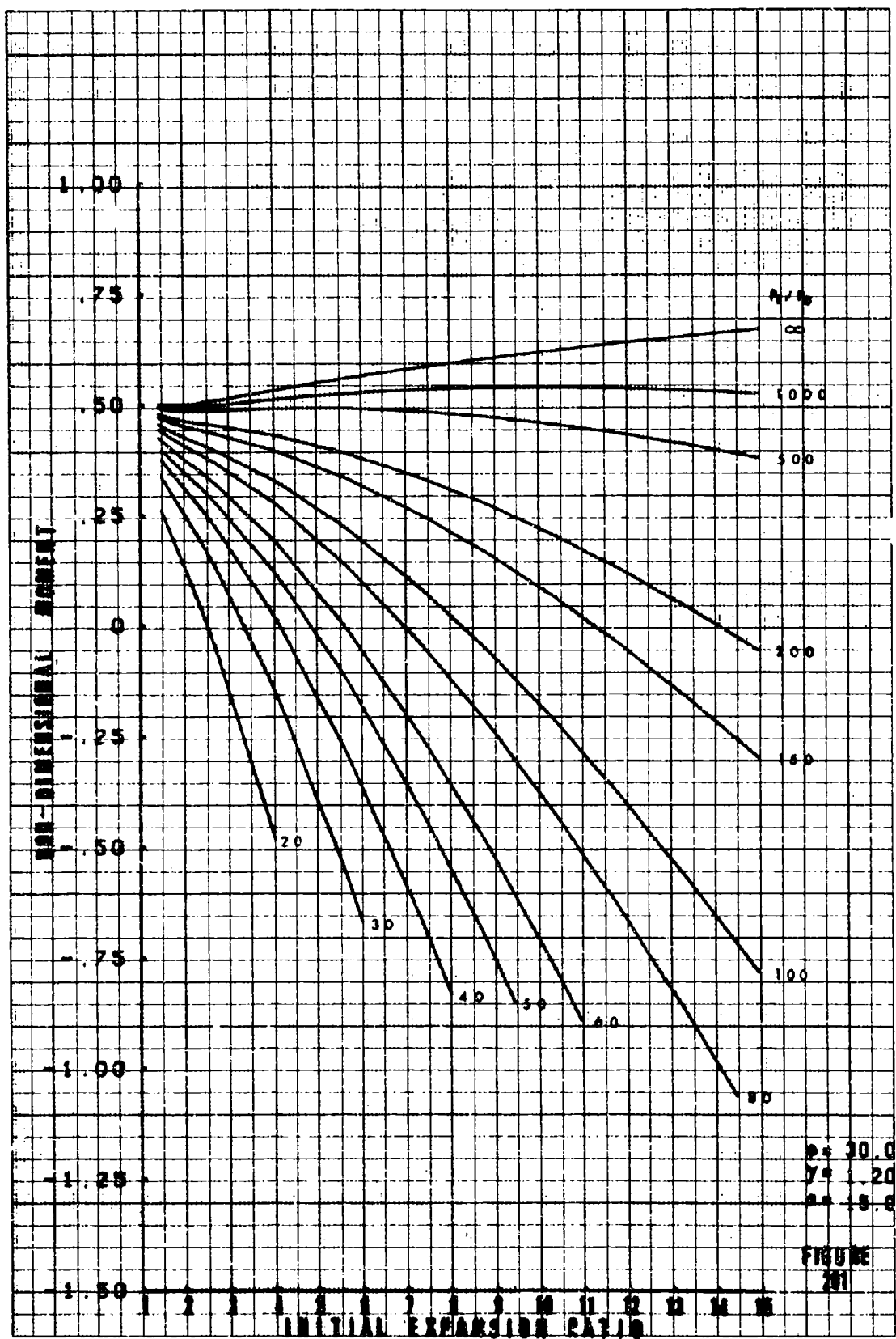
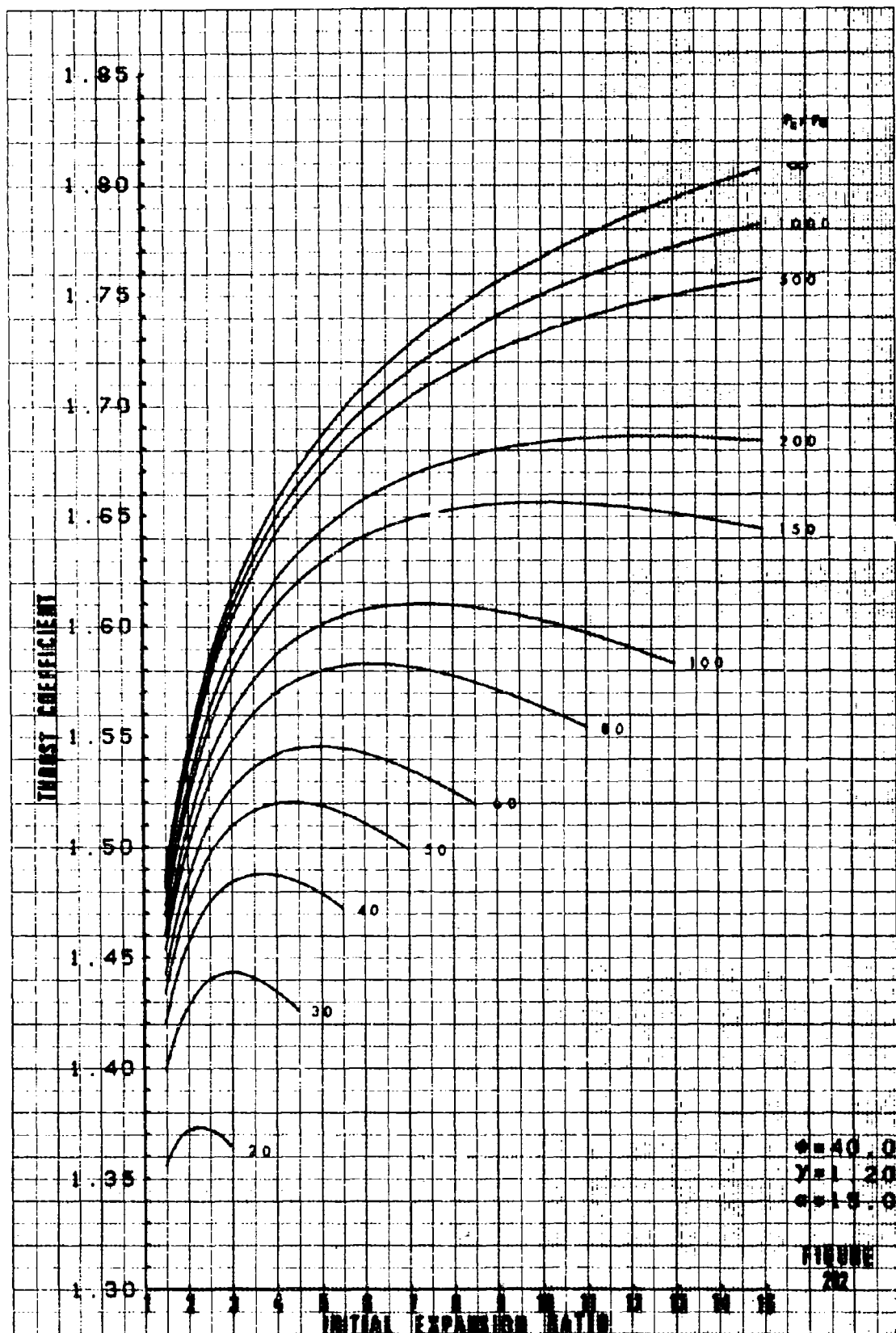


FIGURE 10









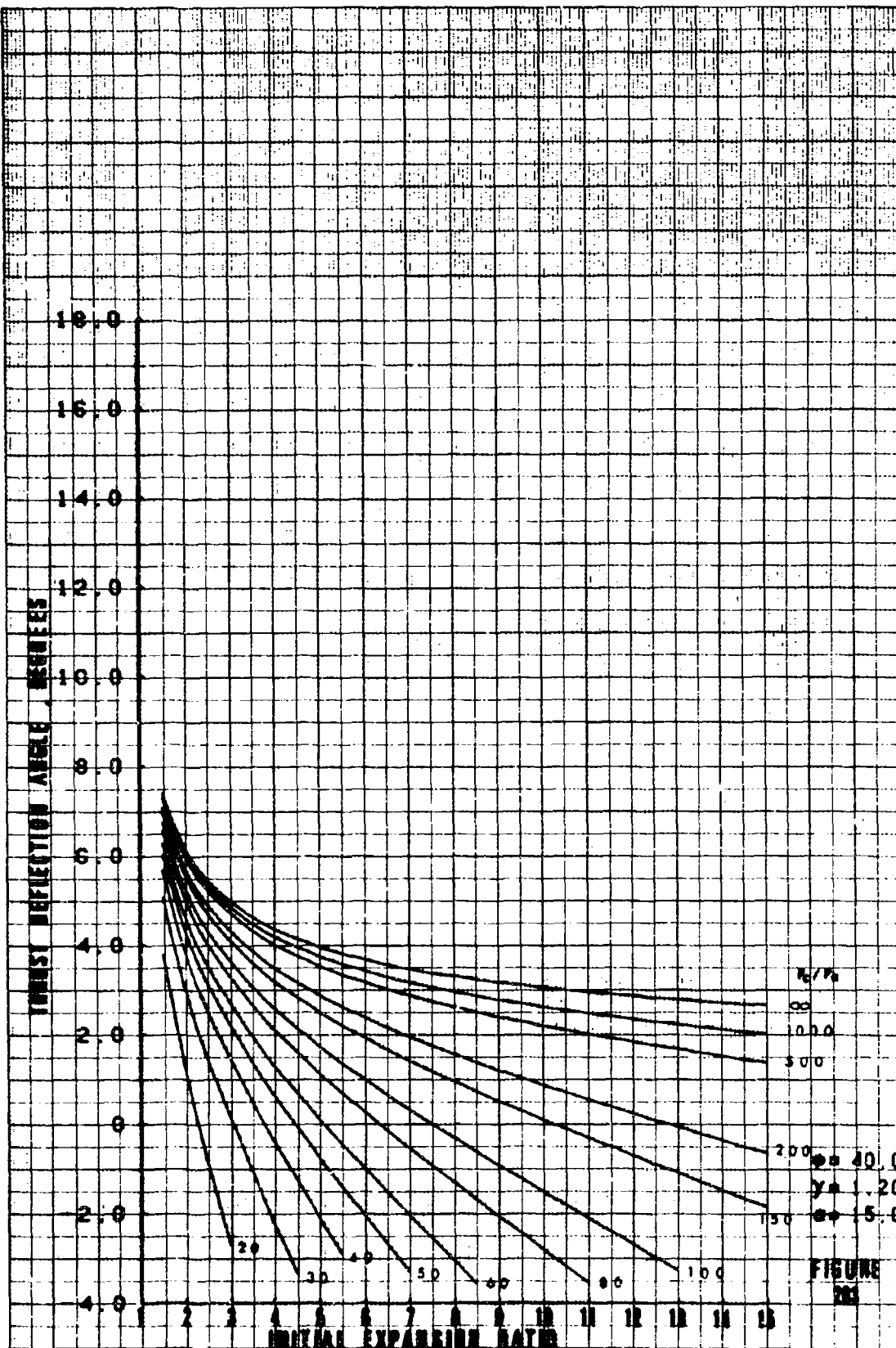
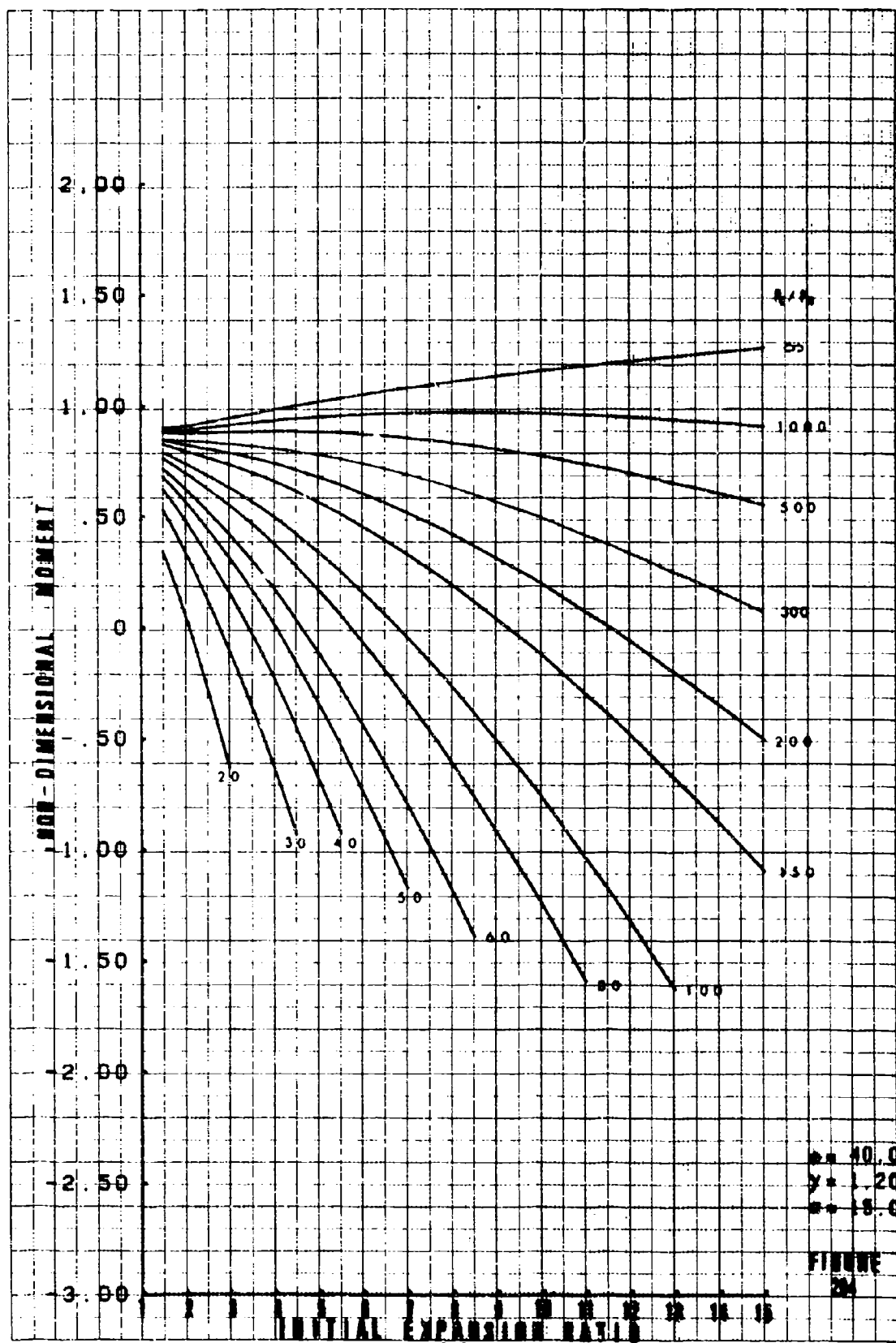
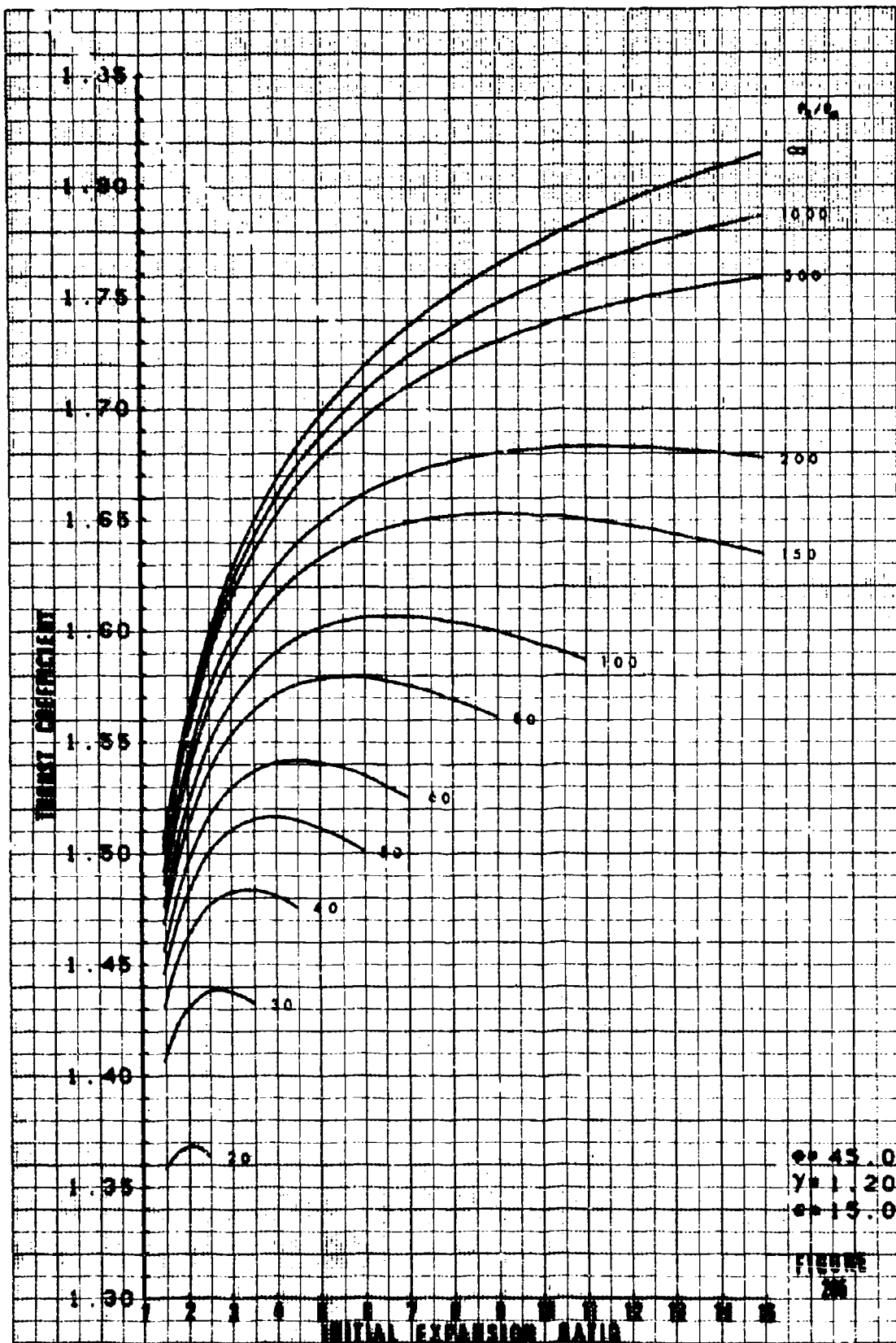
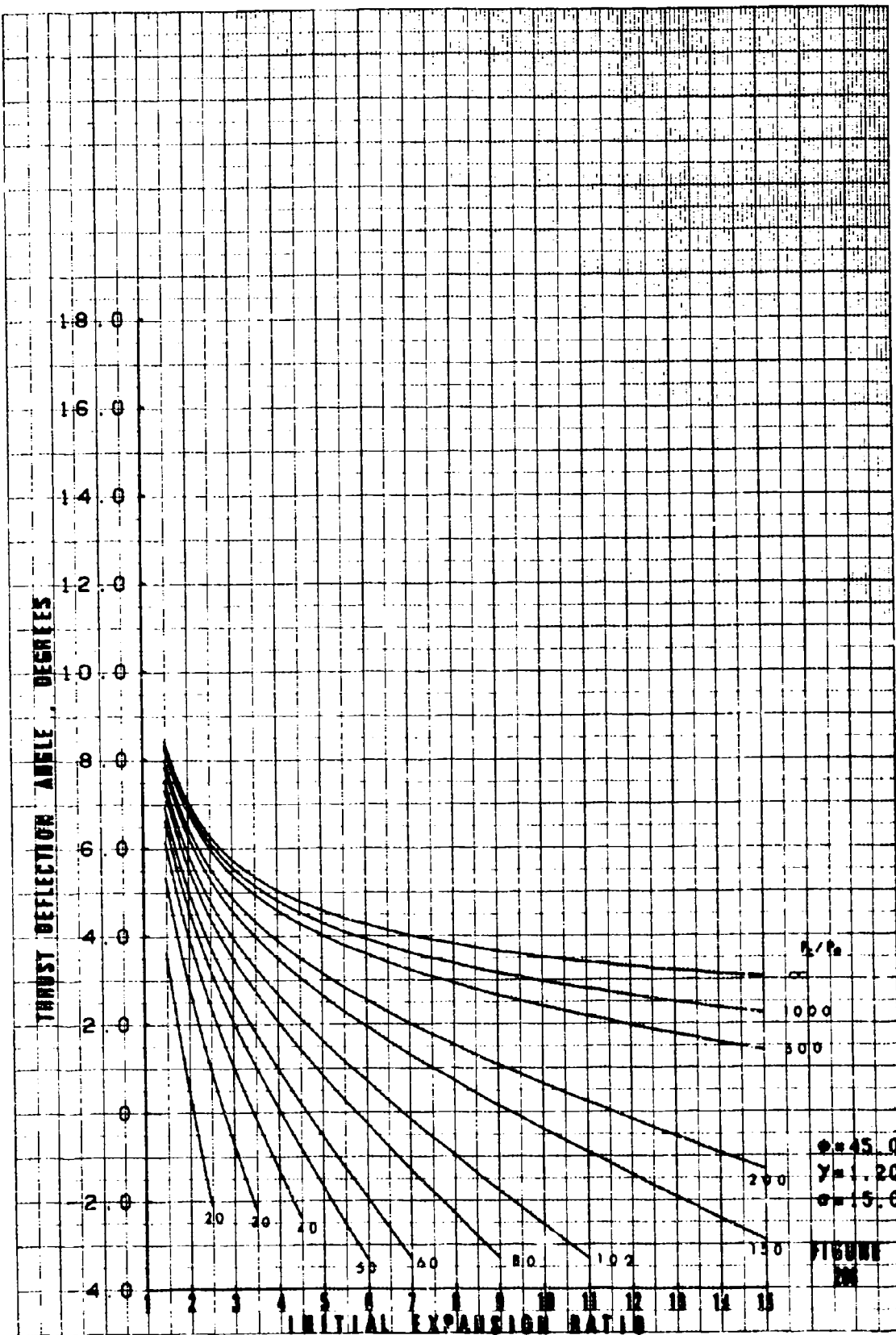
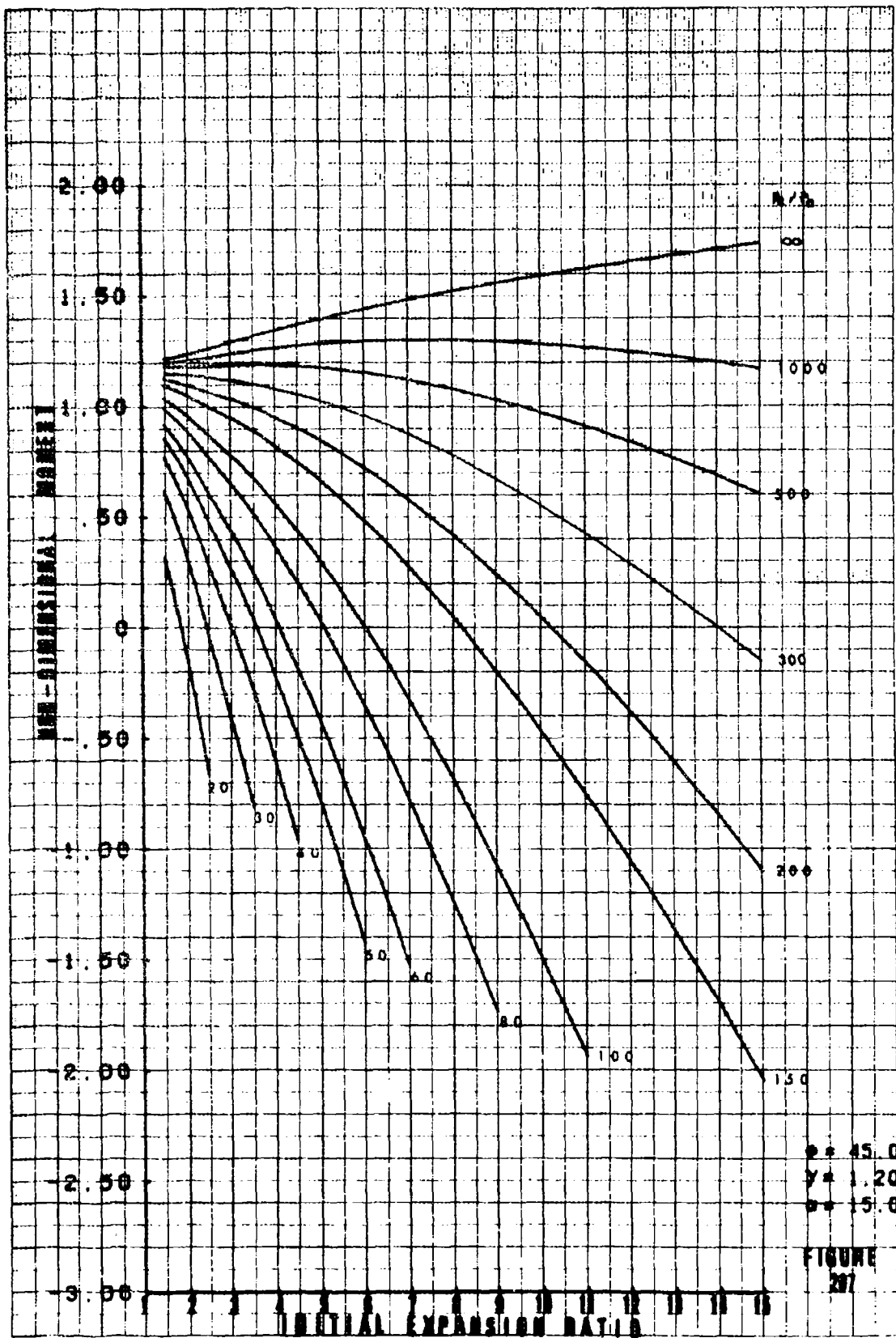


FIGURE 203



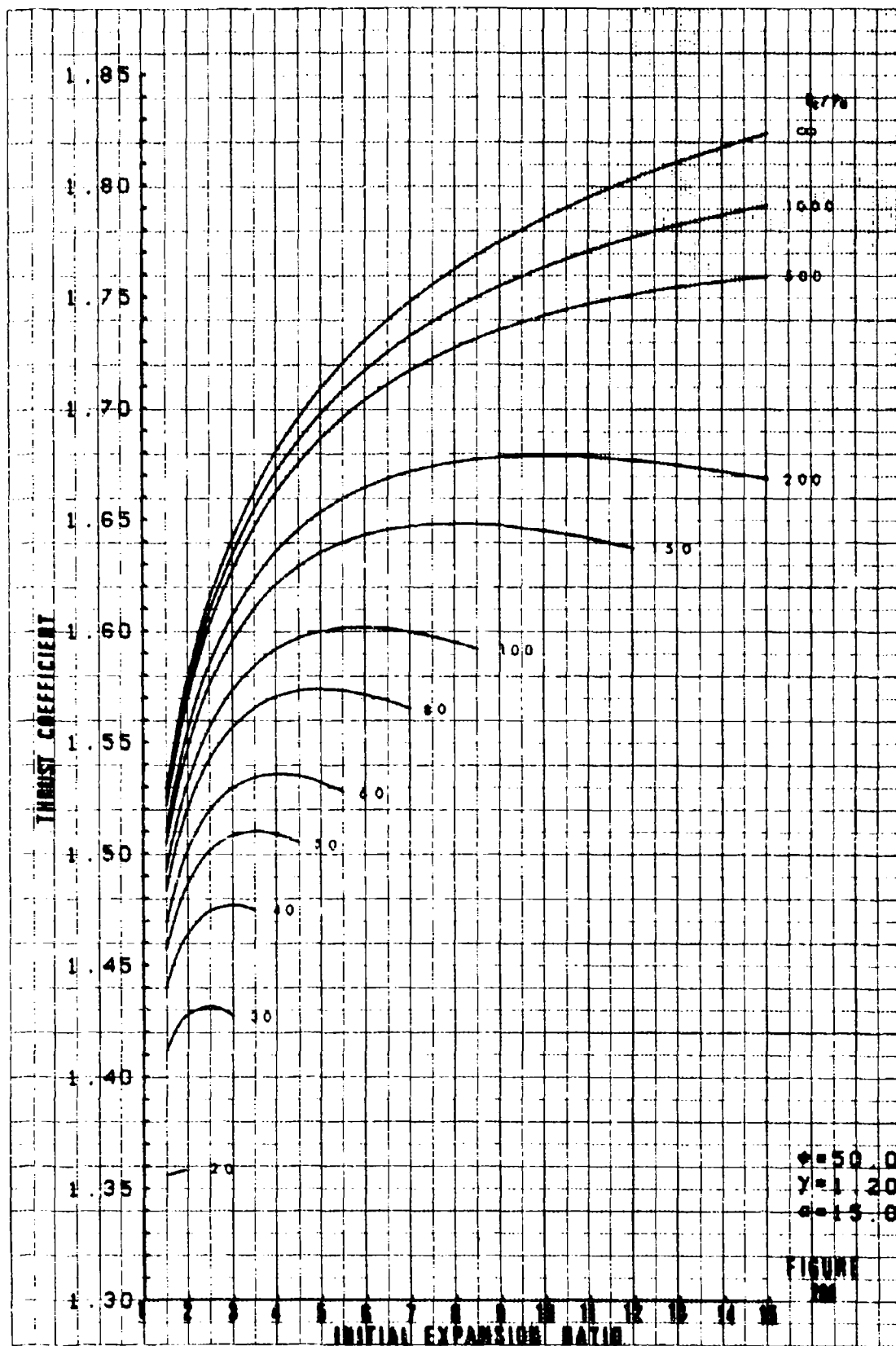






$\mu = 45.0$
 $\gamma = 1.20$
 $\rho = 15.0$

FIGURE 207



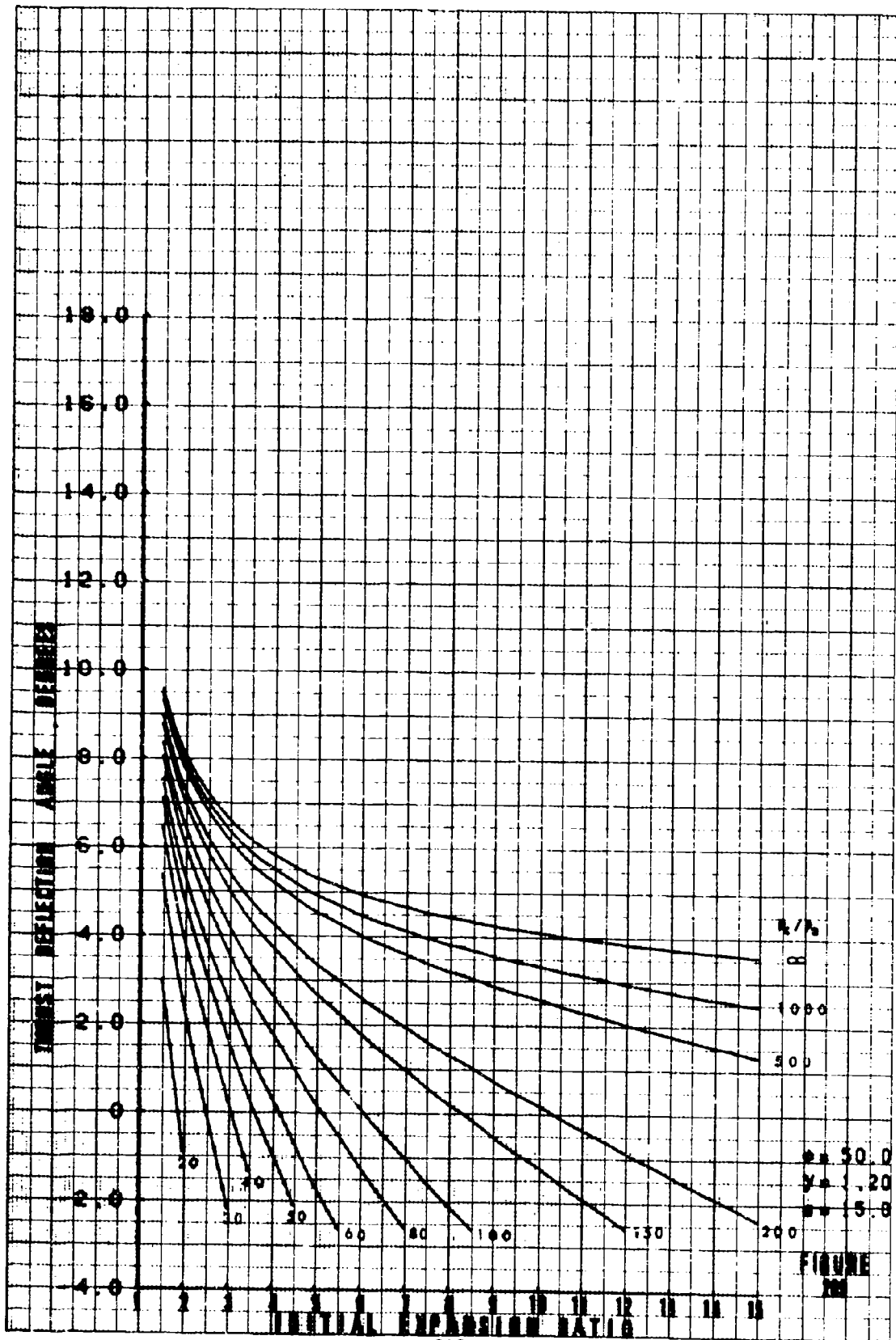
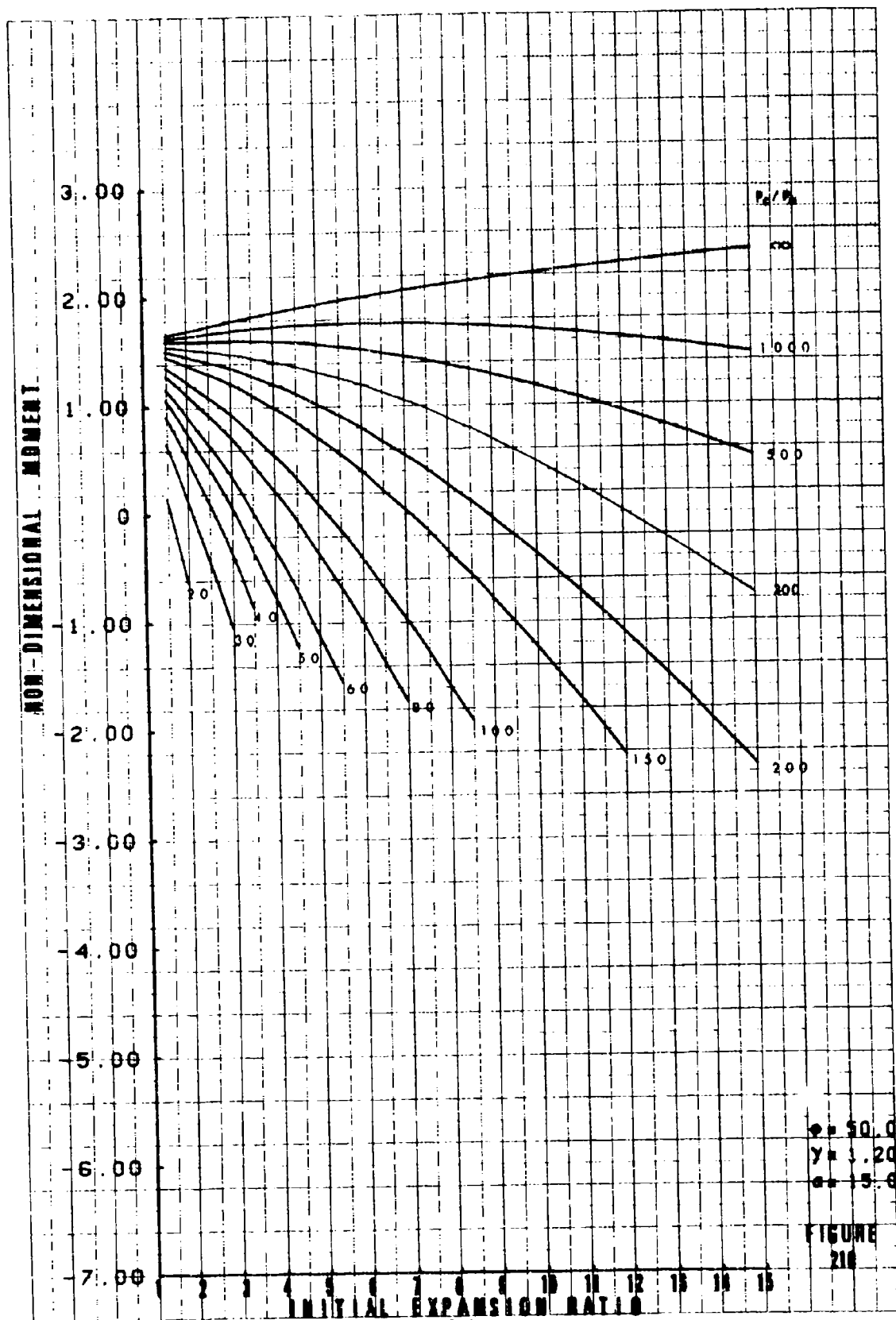


FIGURE 20



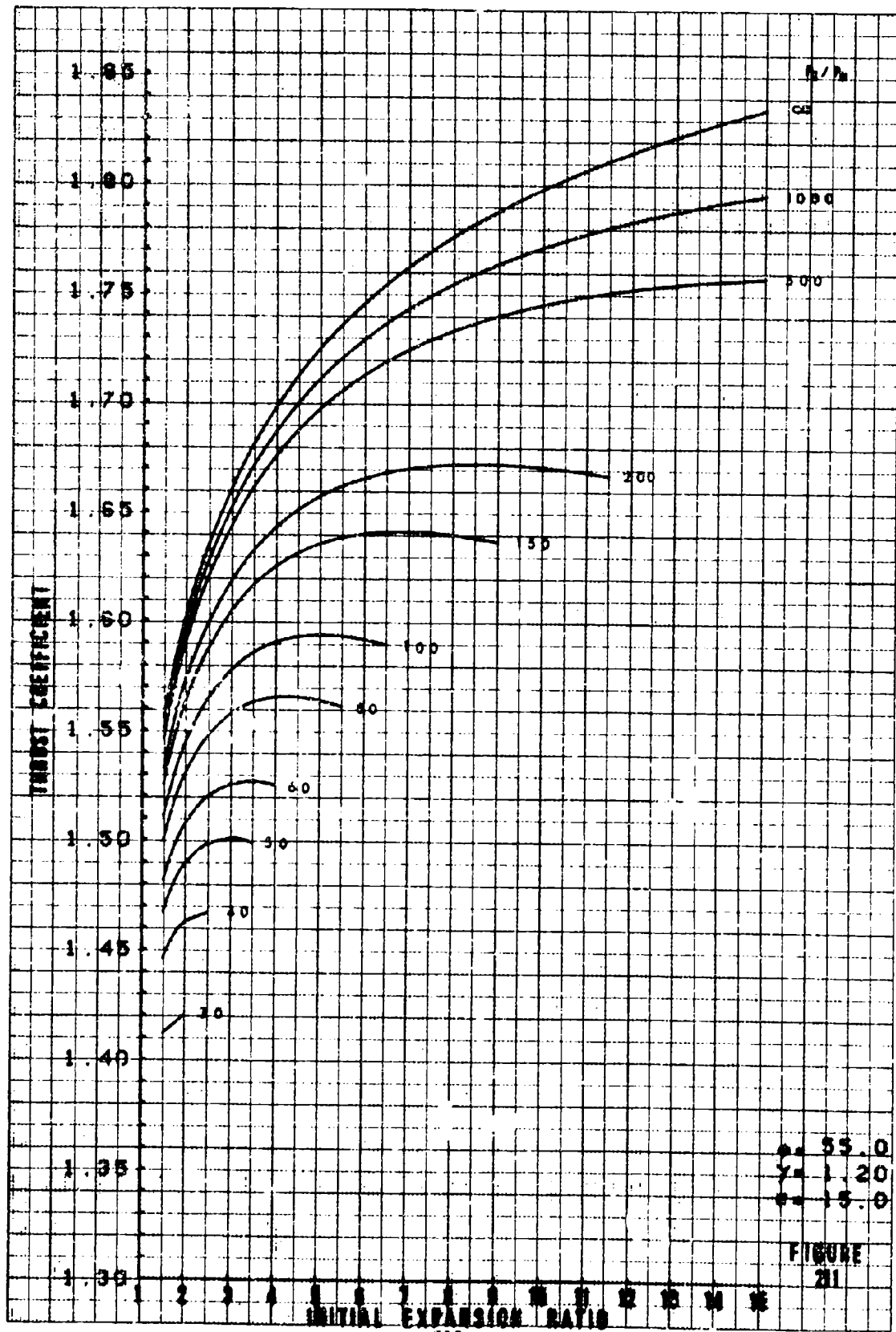


FIGURE 211

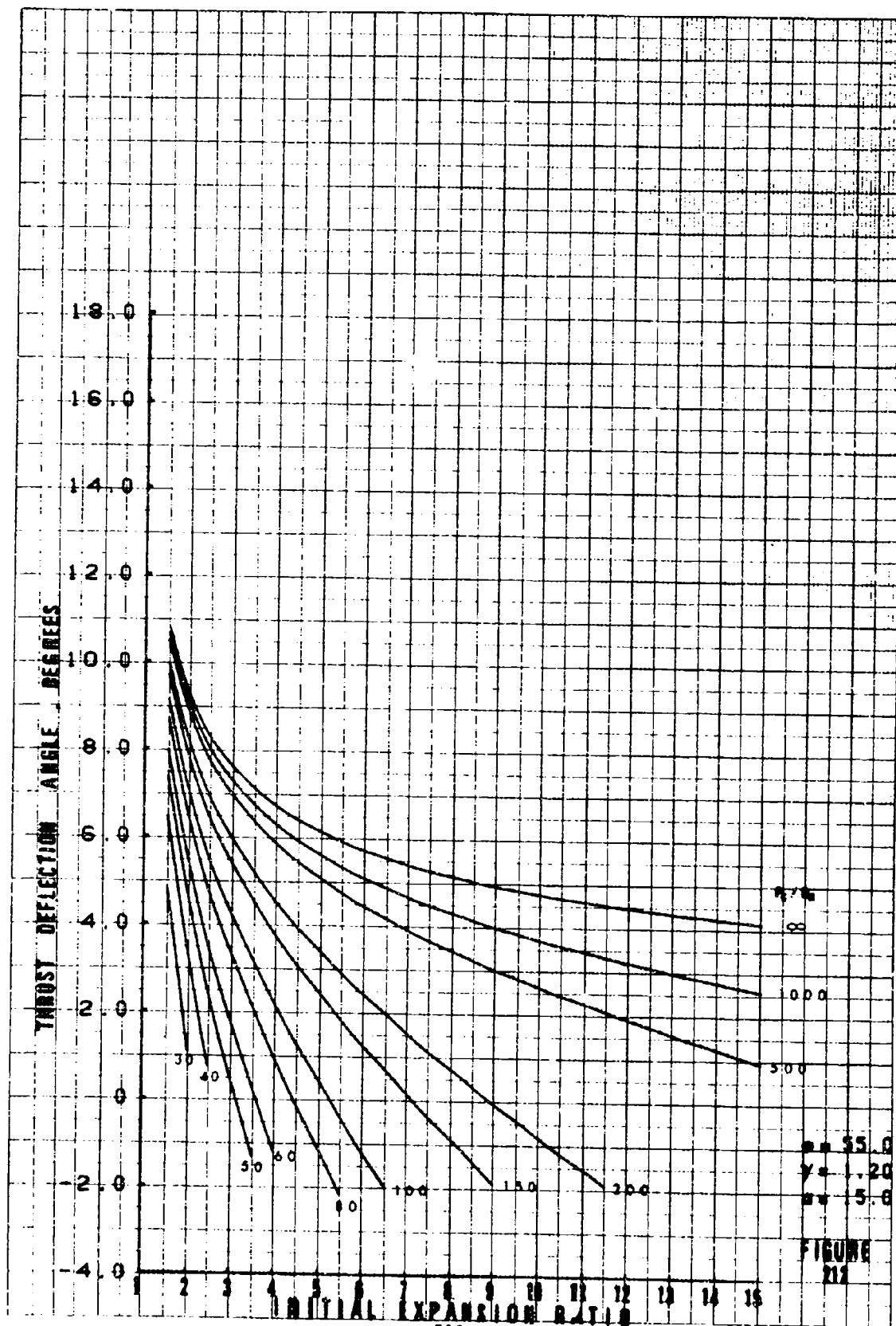
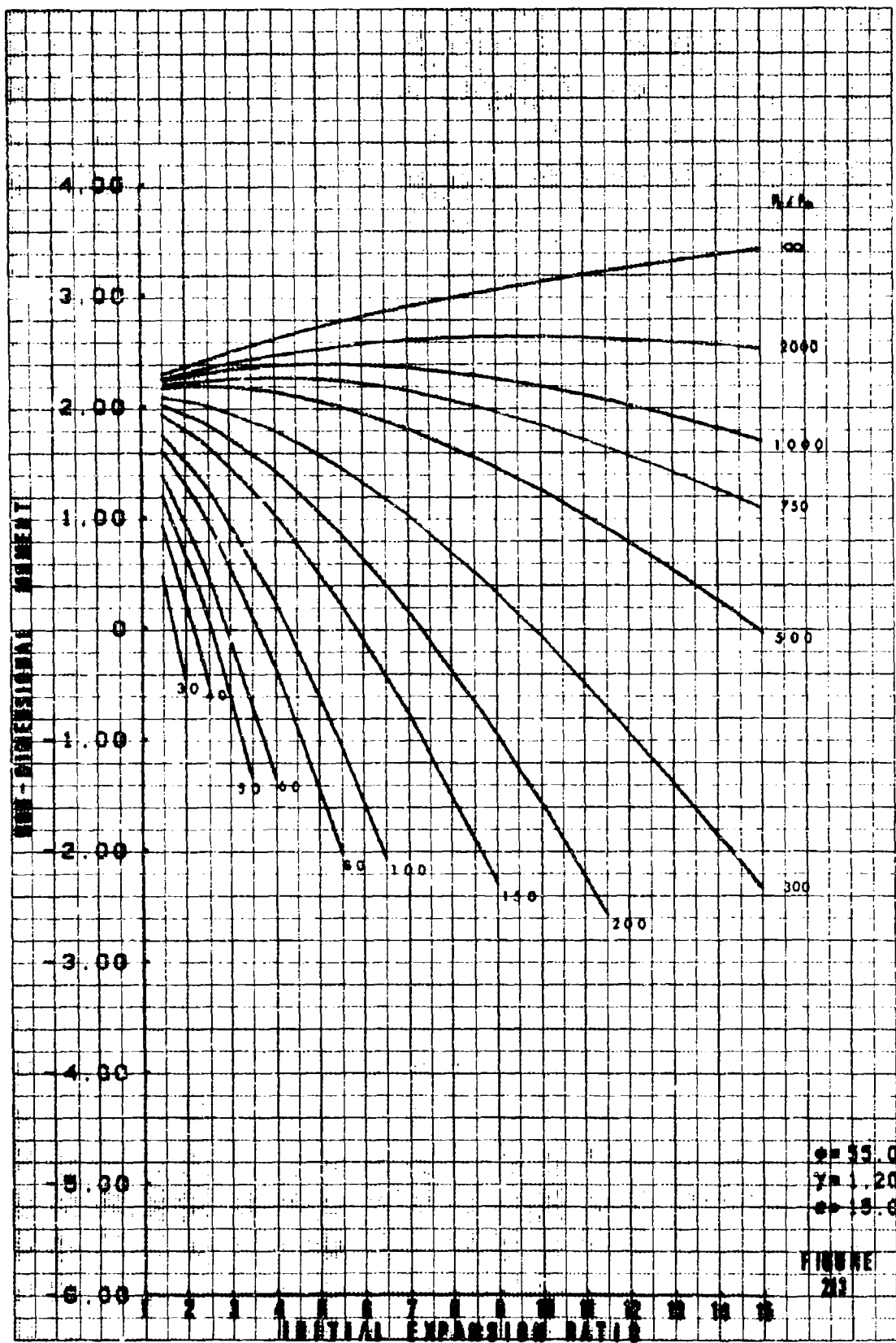


FIGURE 212



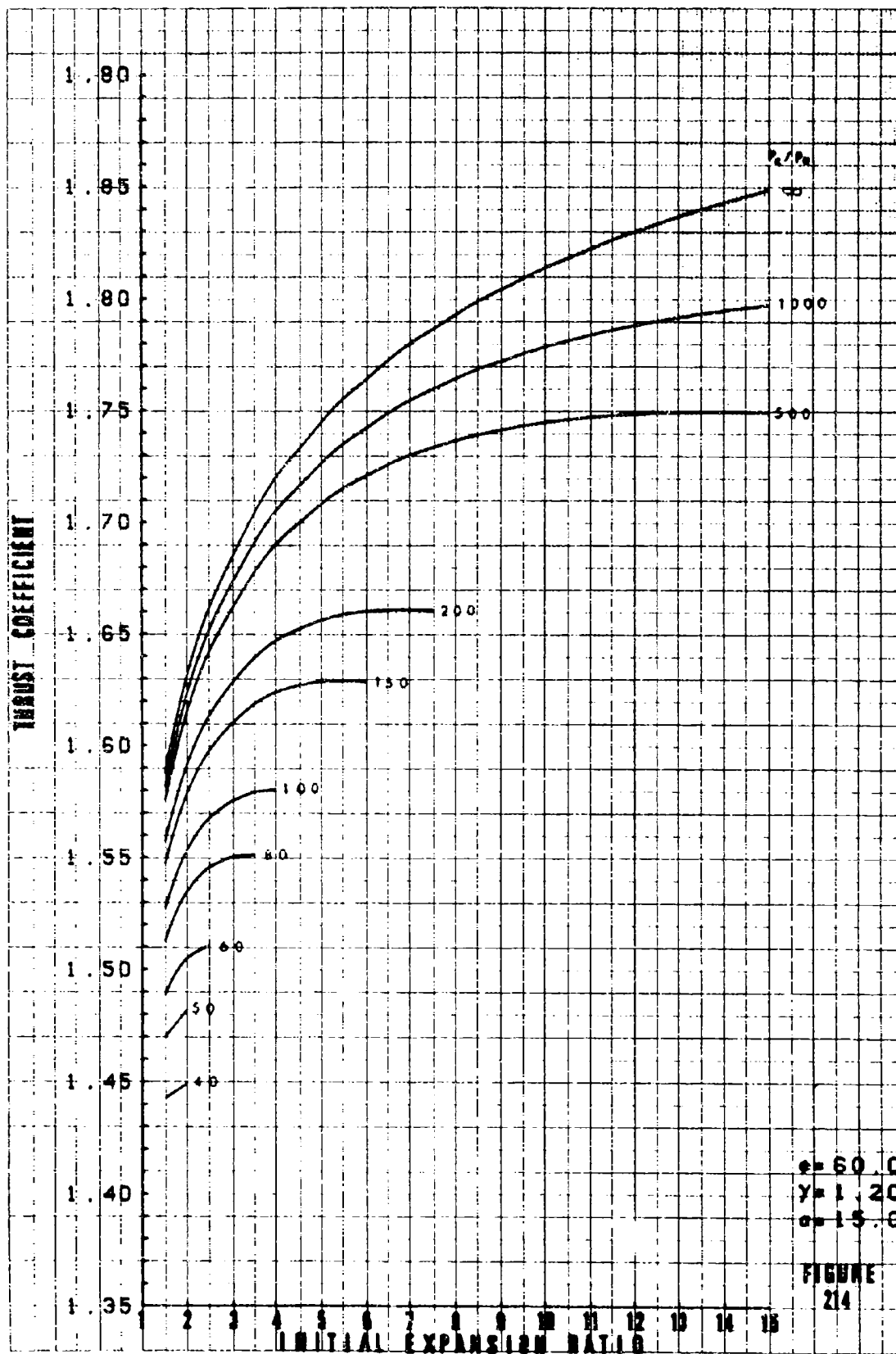


FIGURE
214

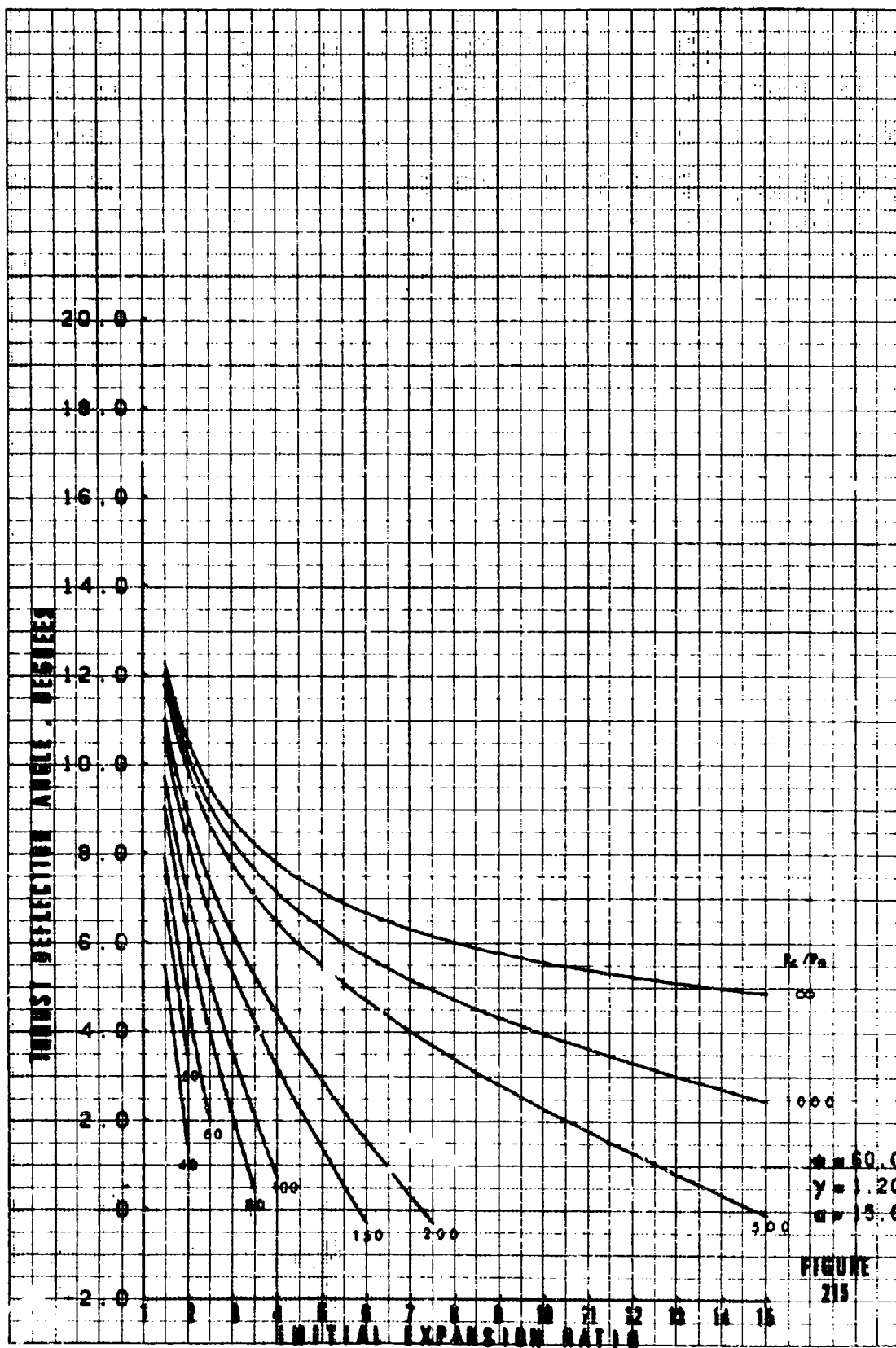
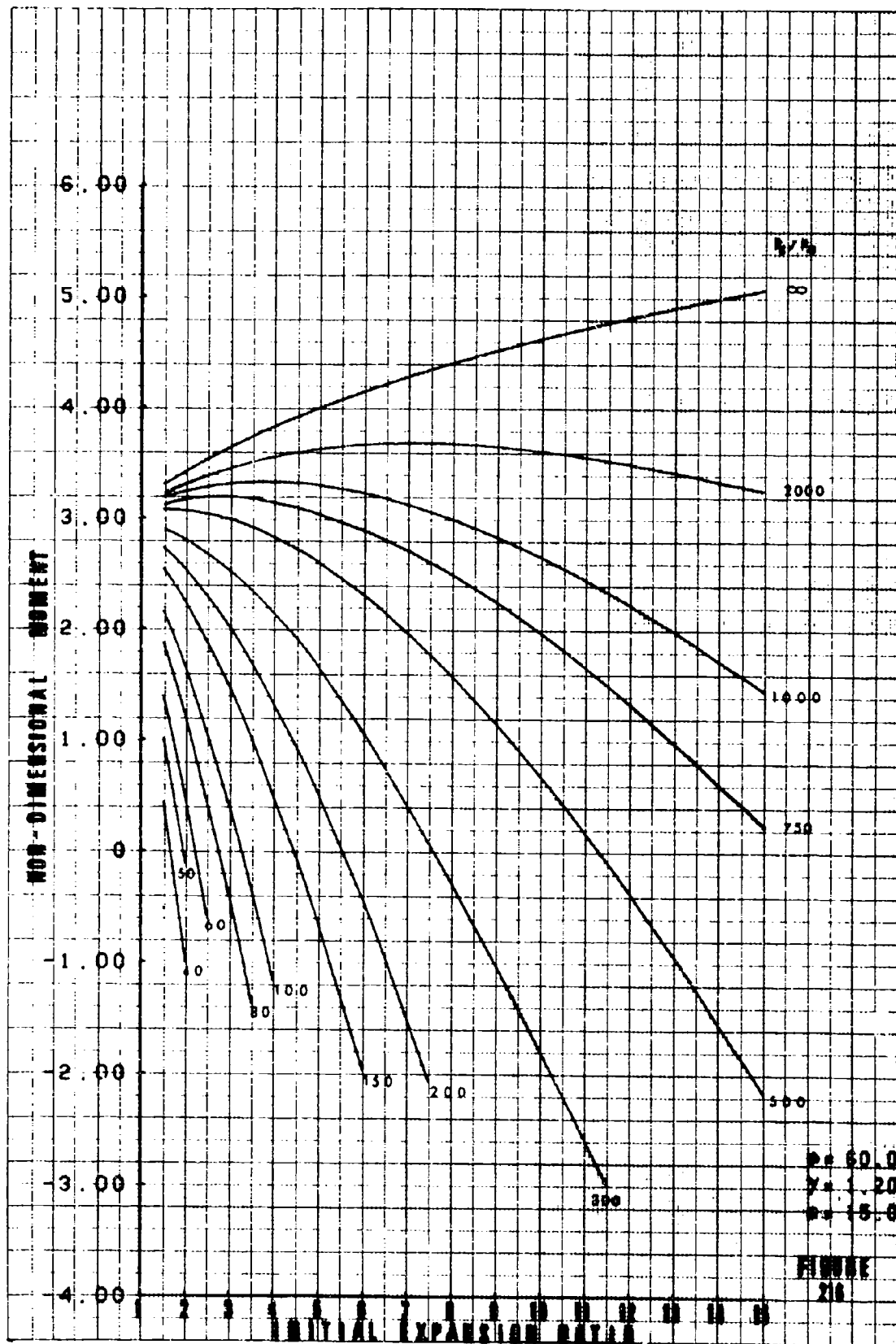
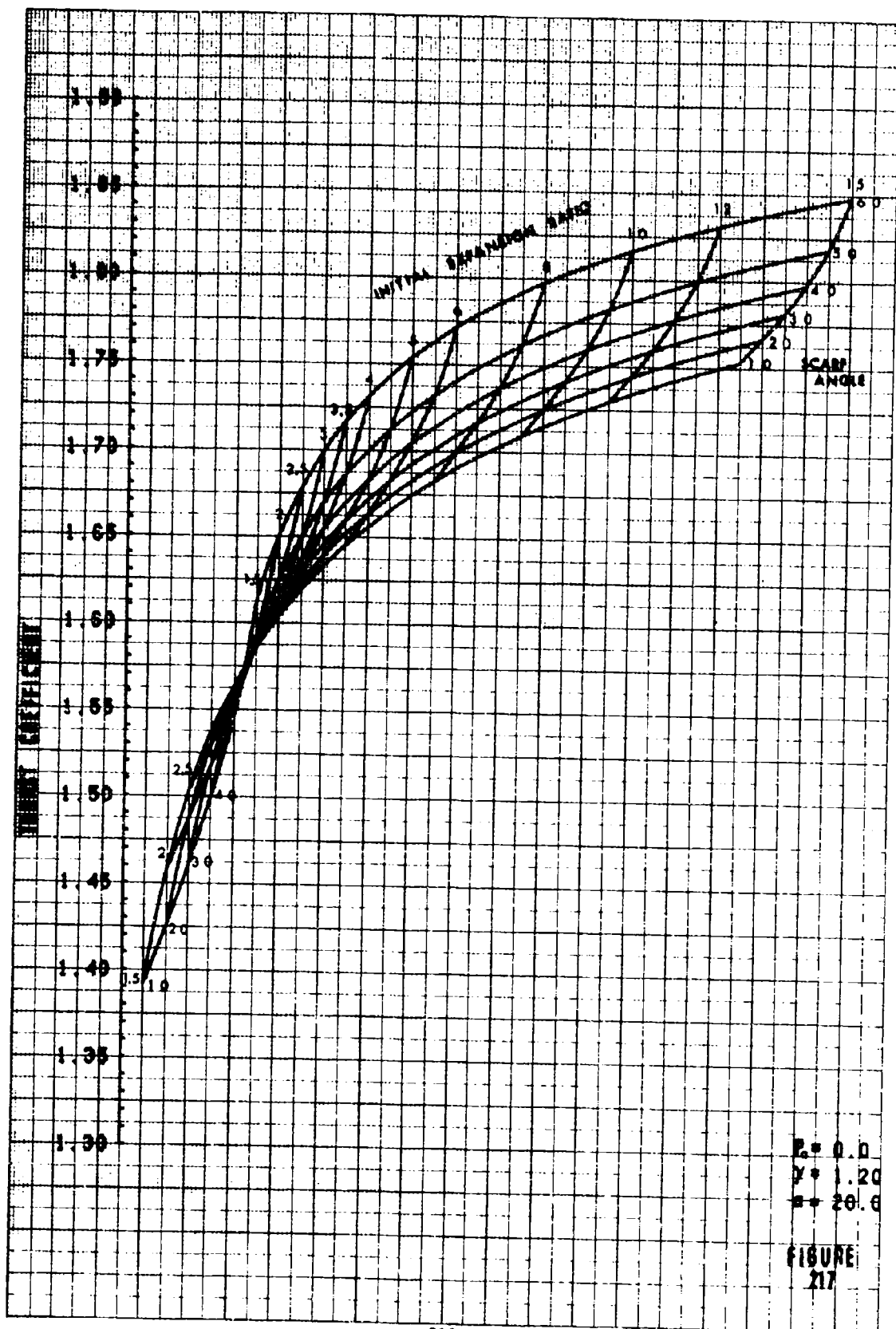
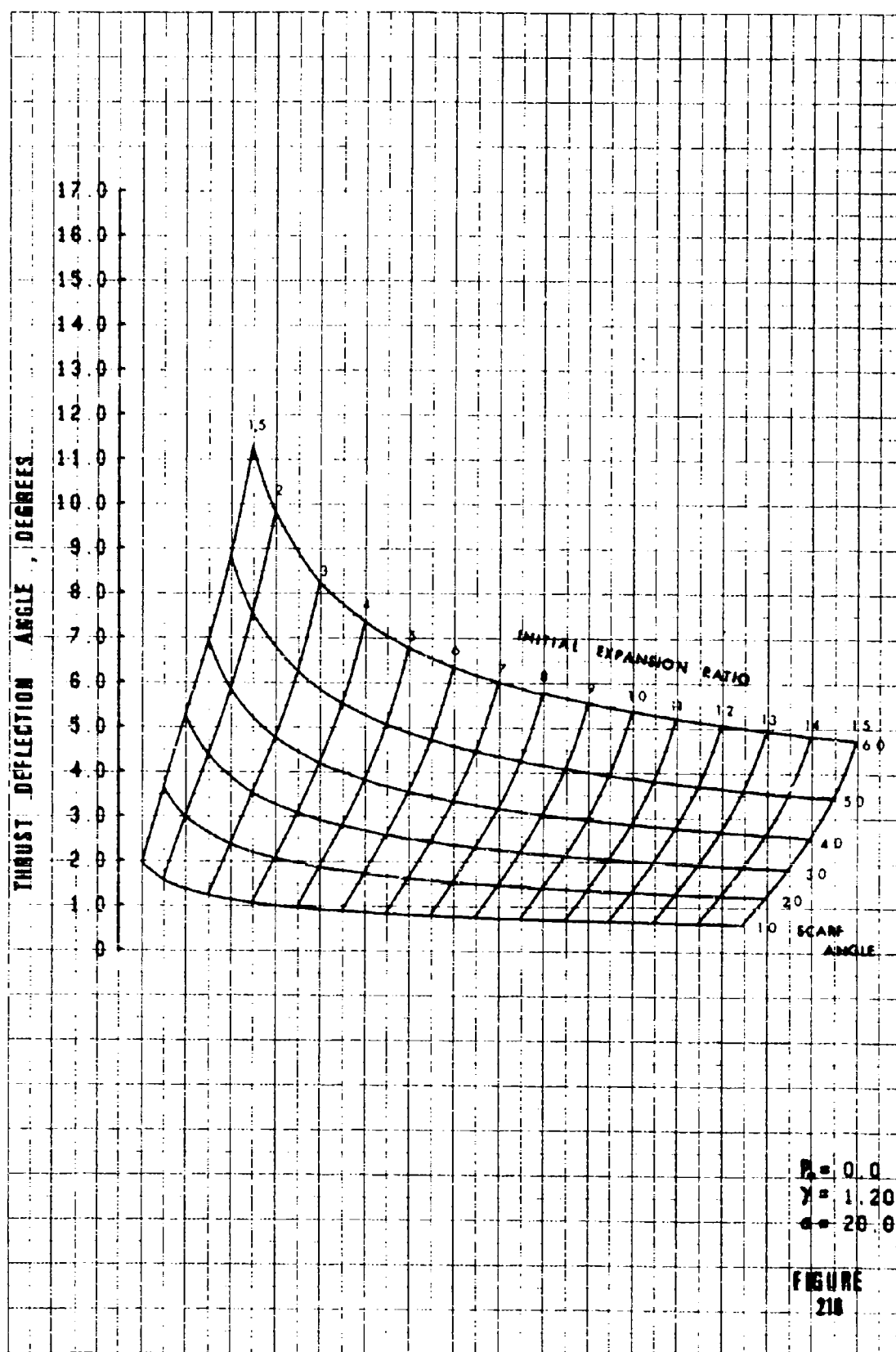
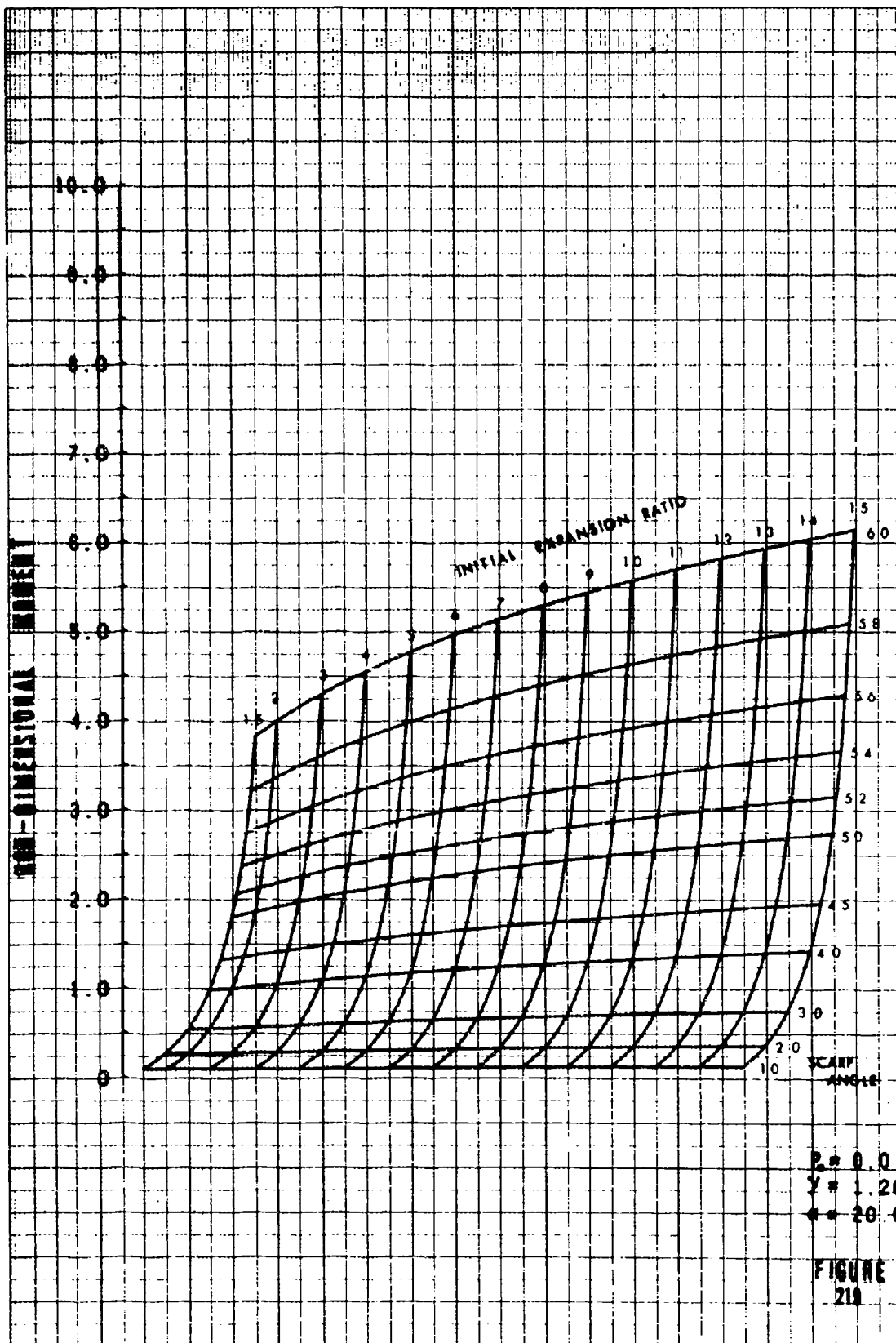


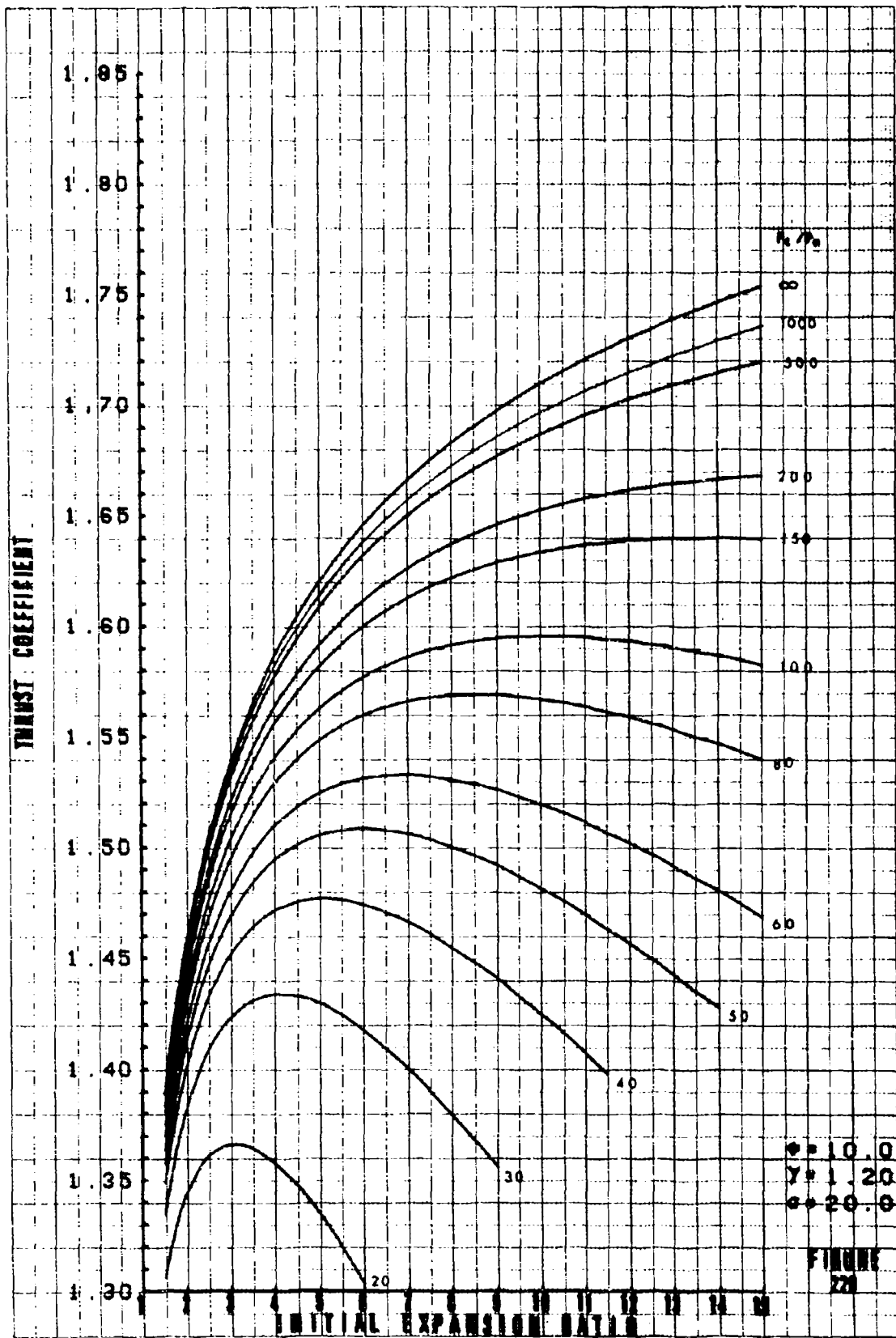
FIGURE
215

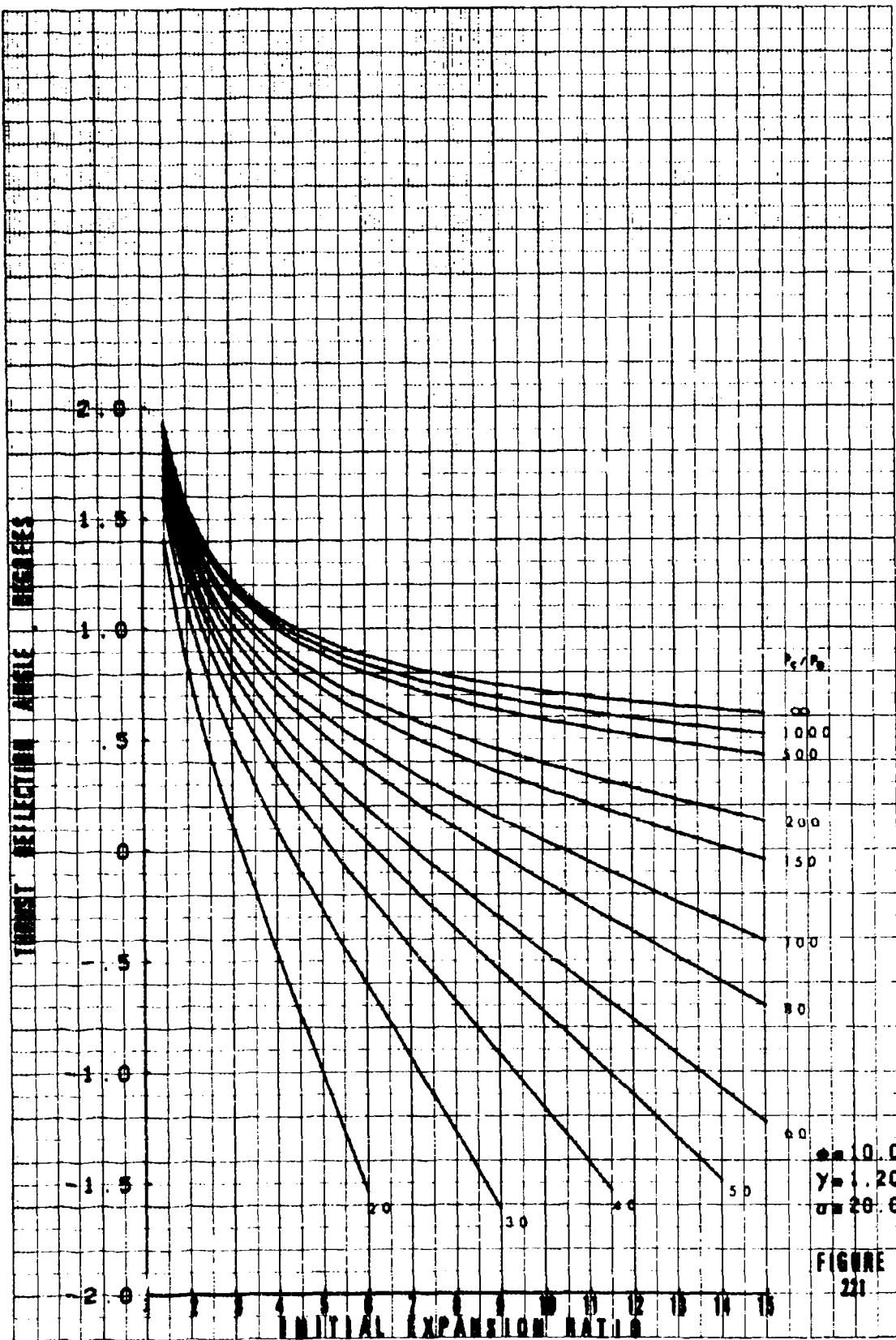


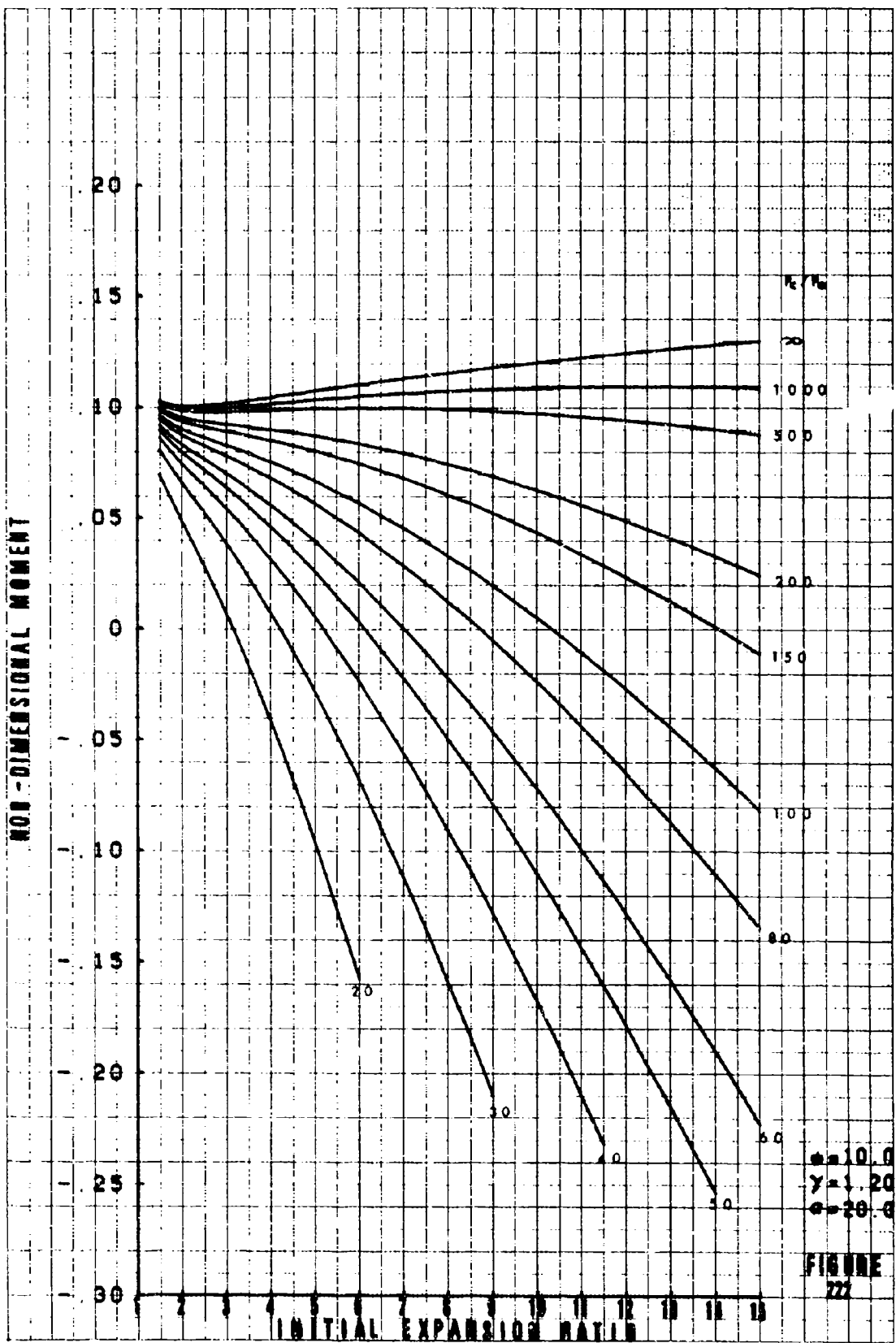


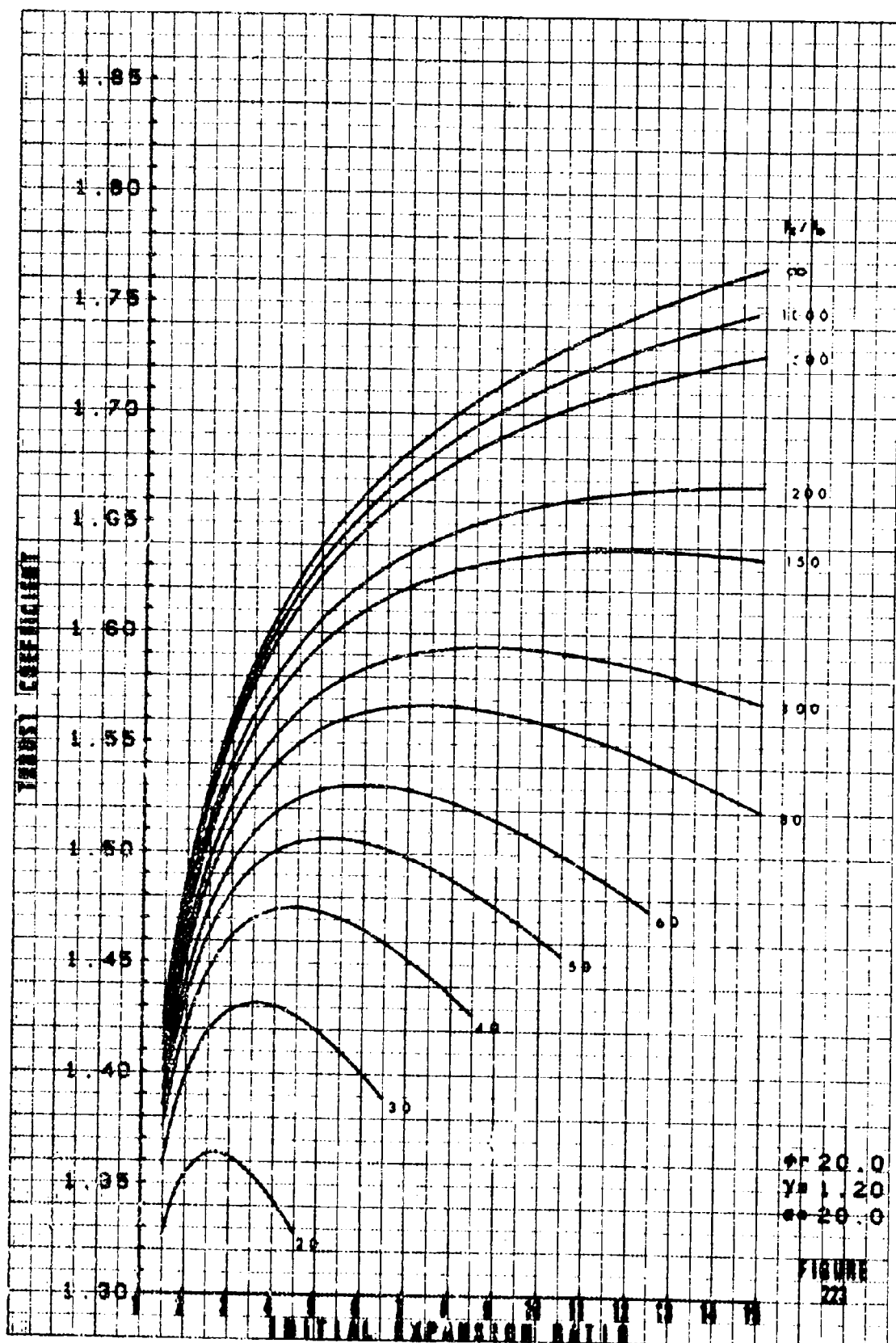


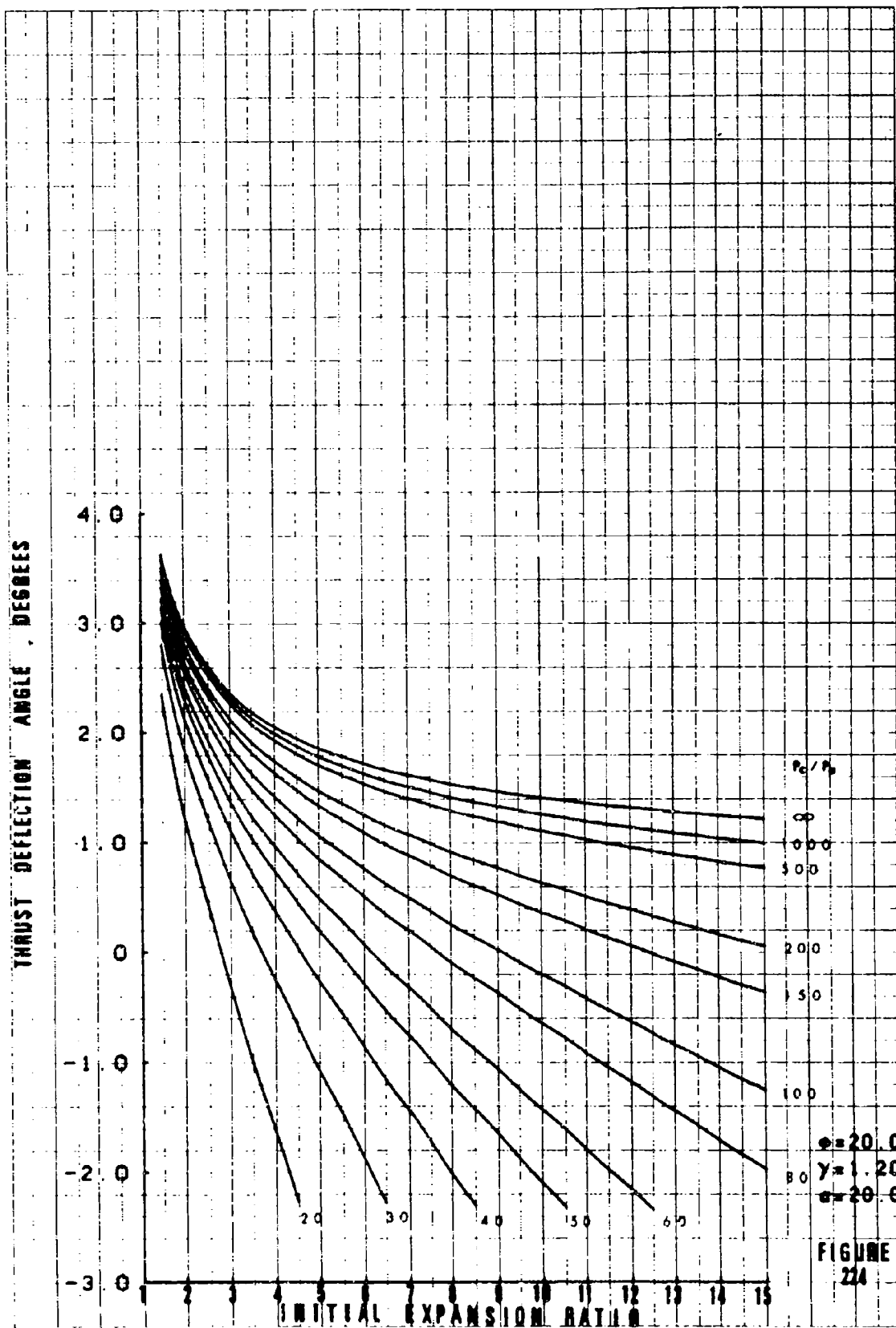


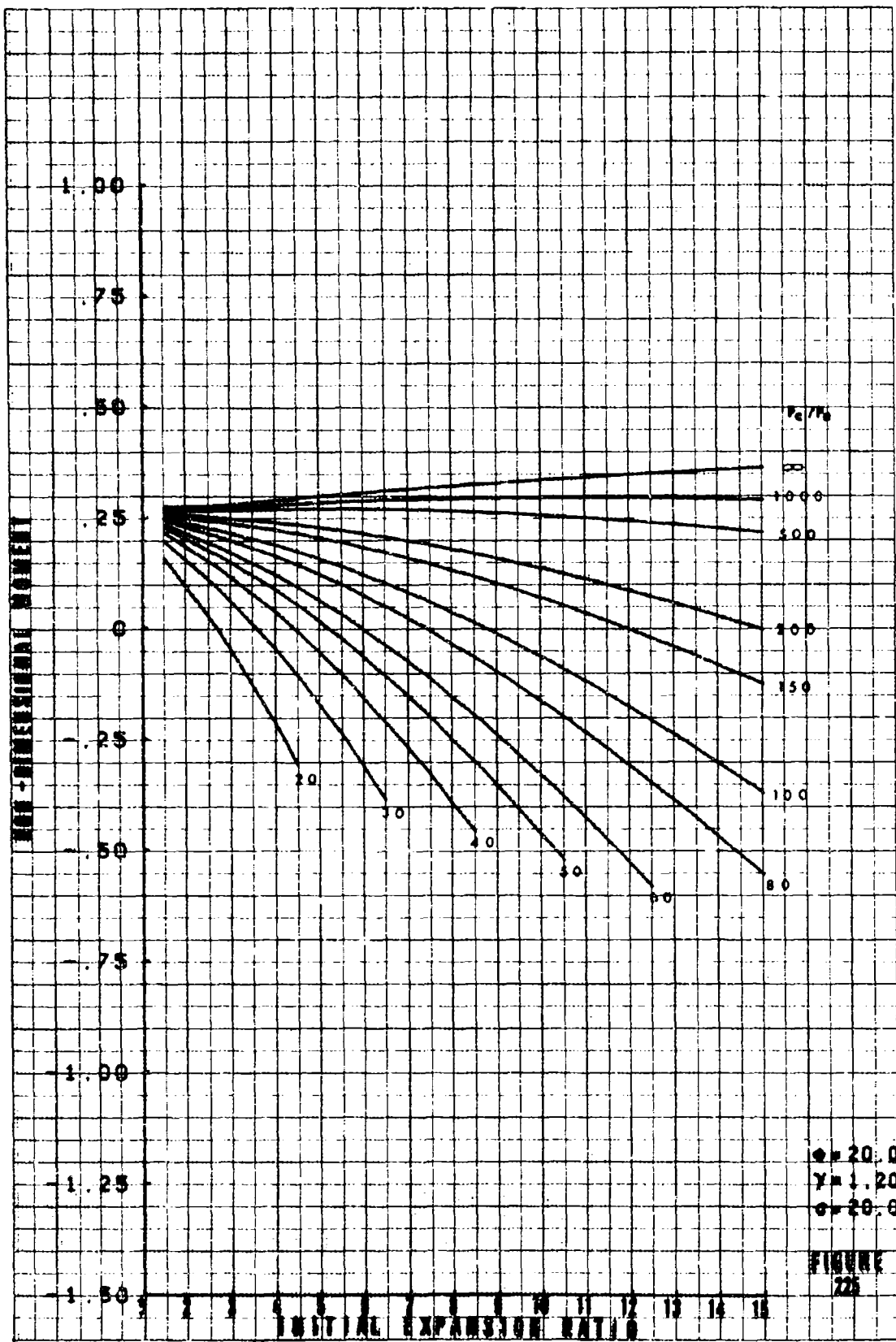












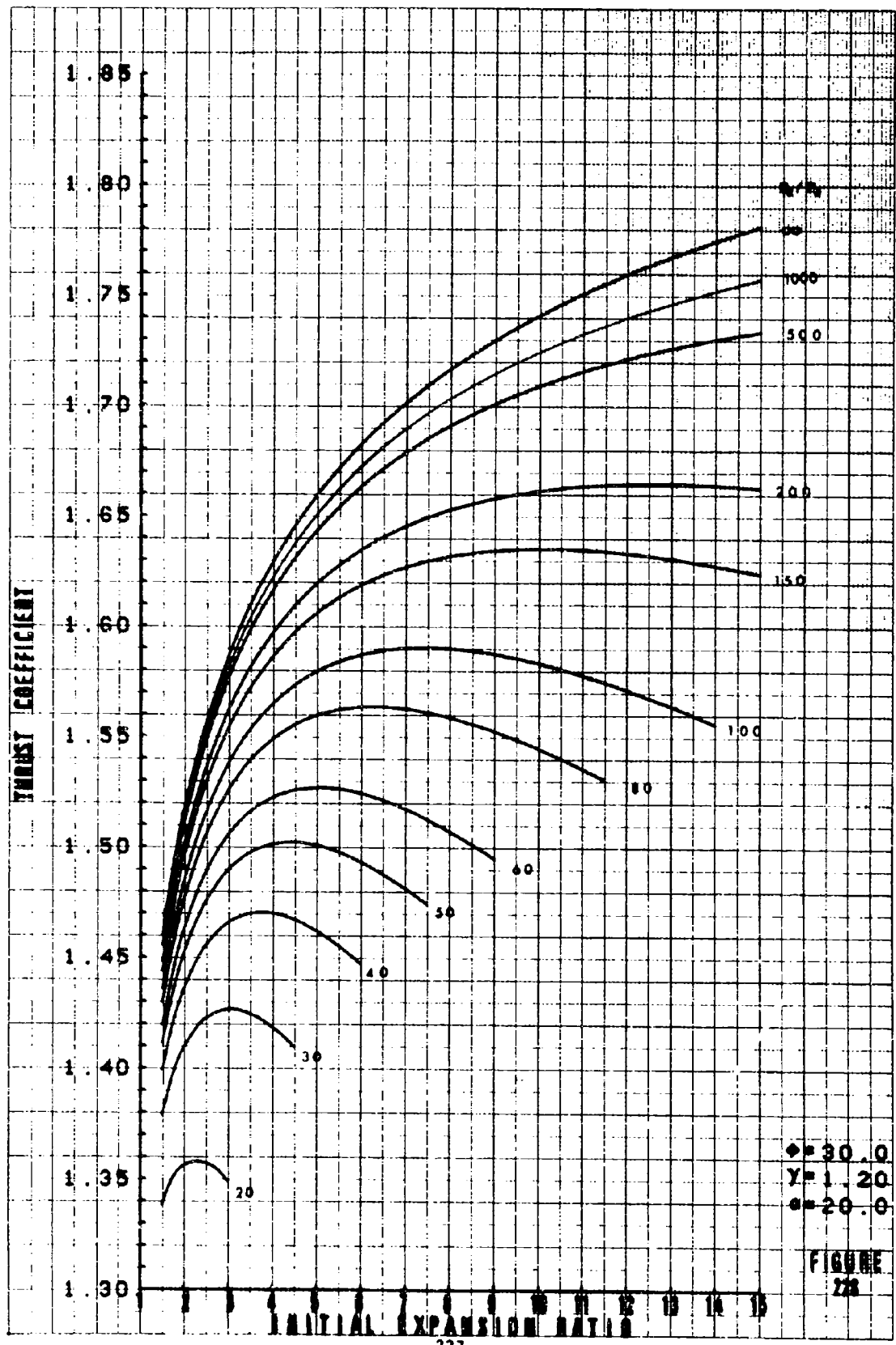
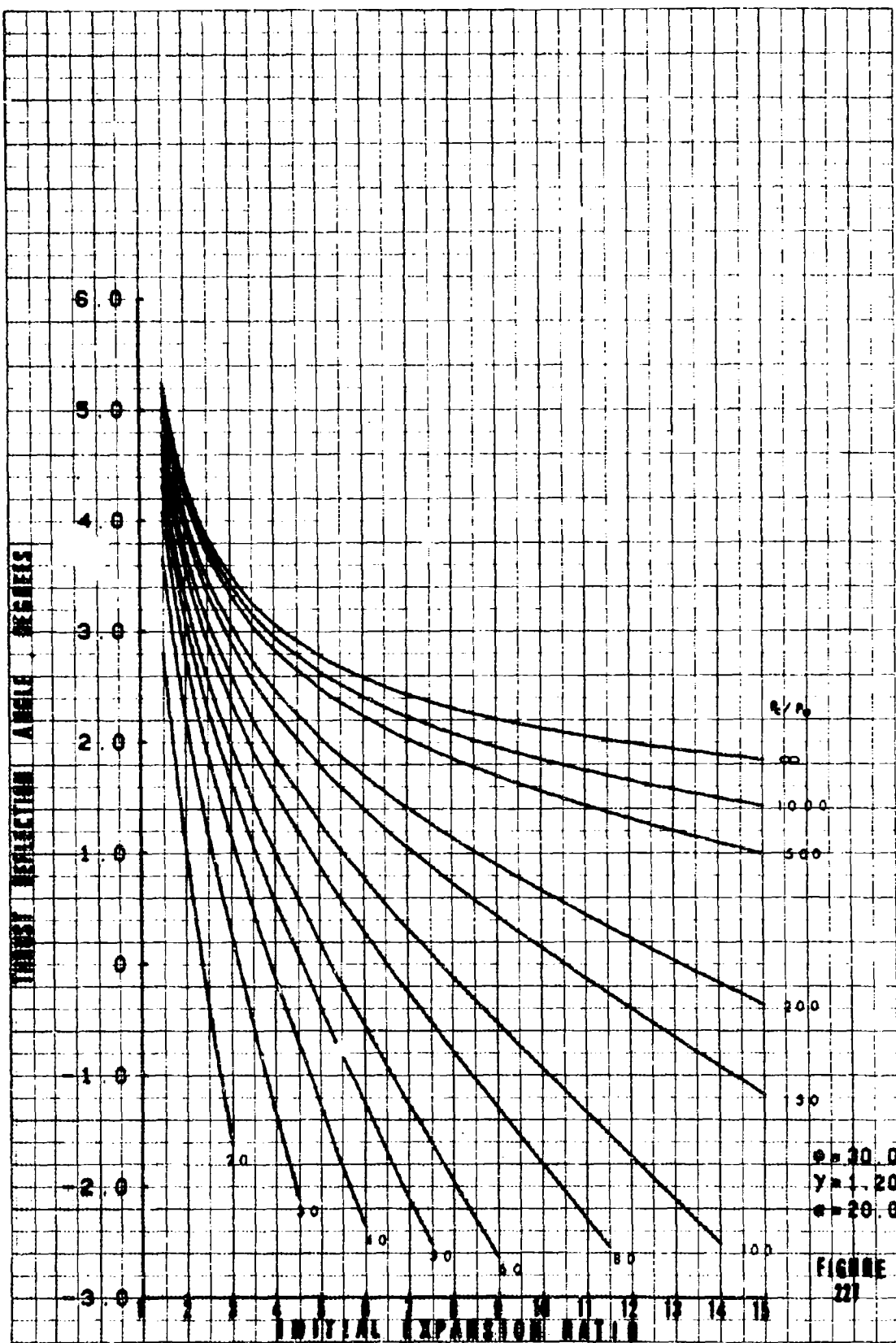
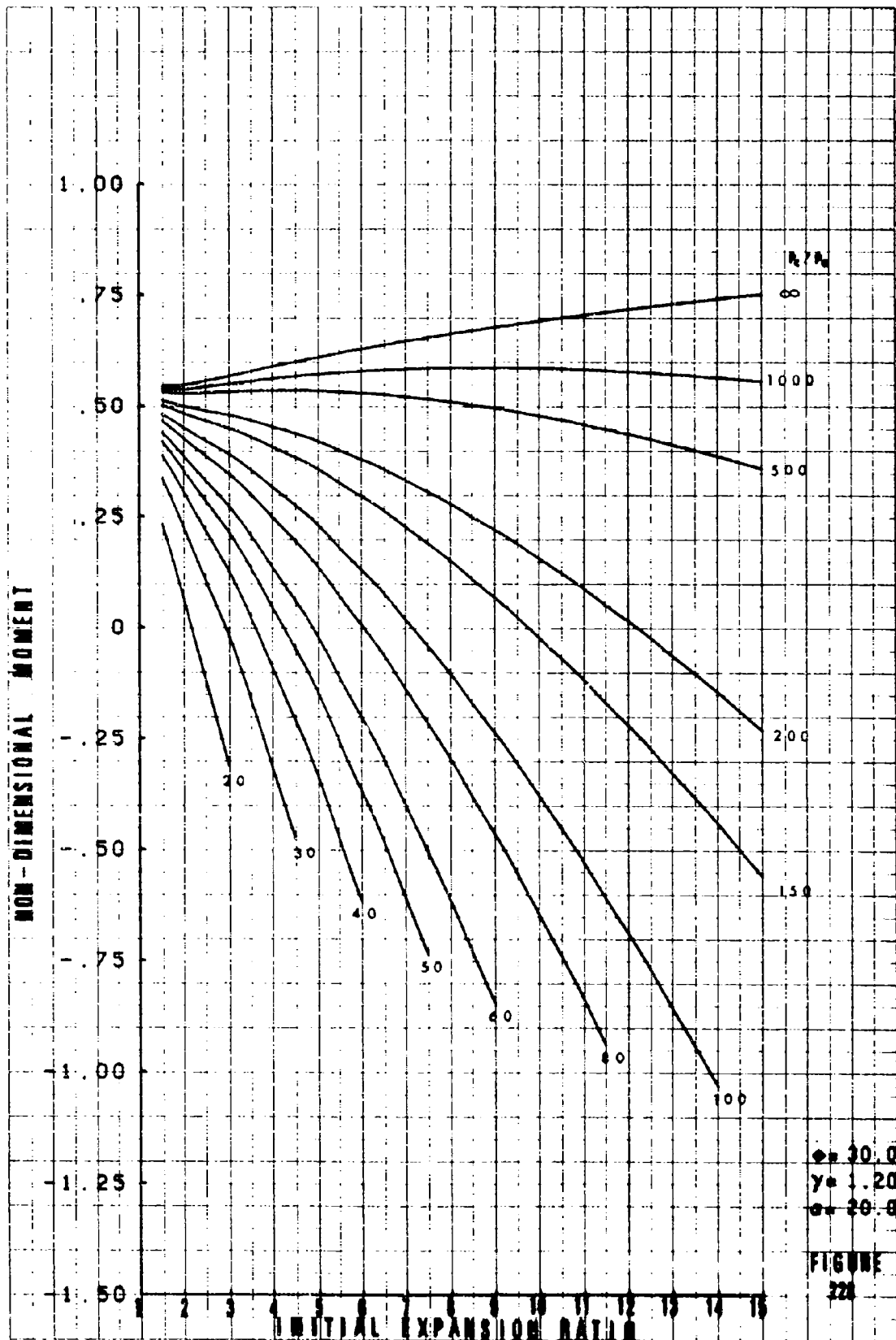
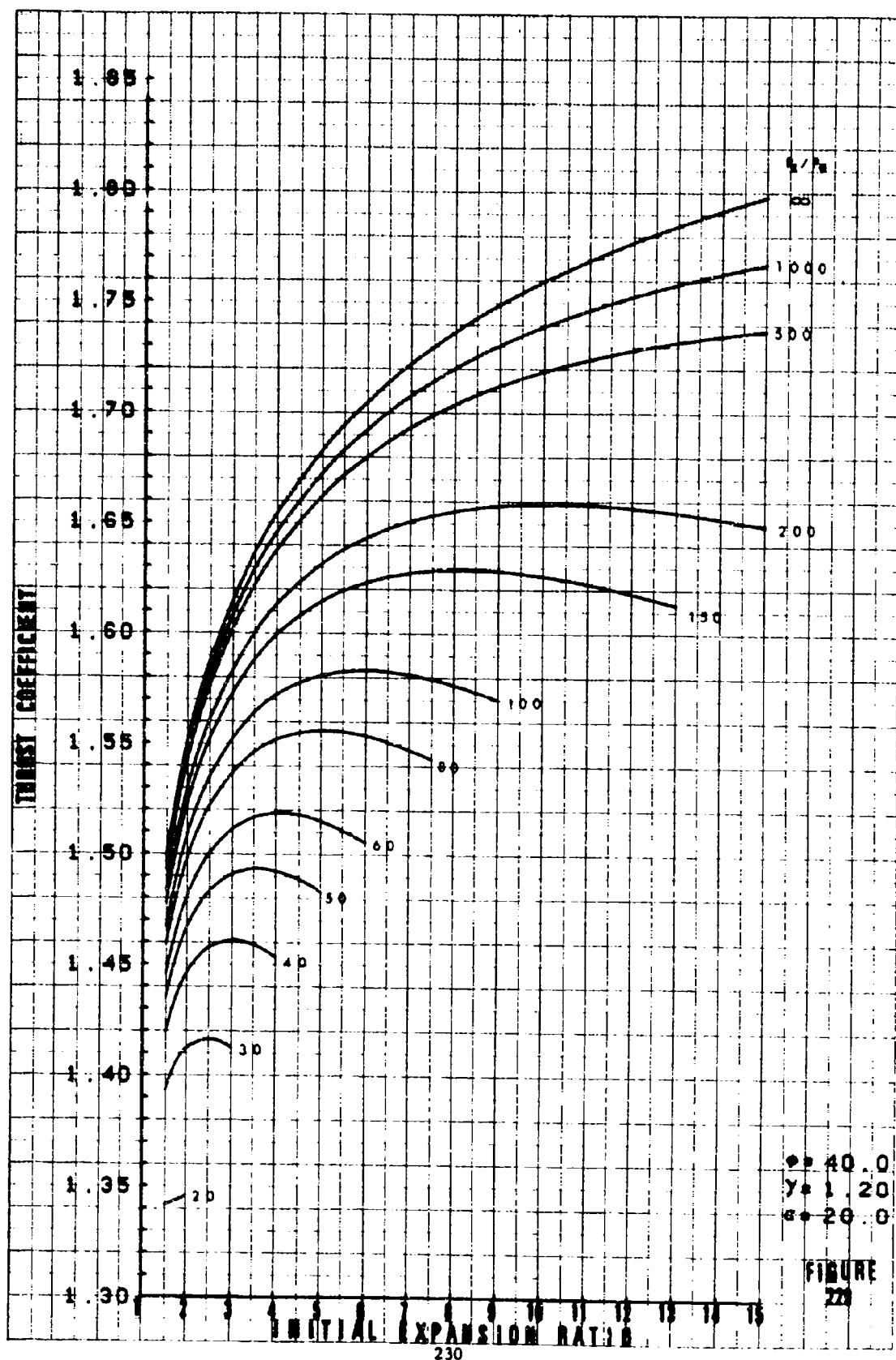
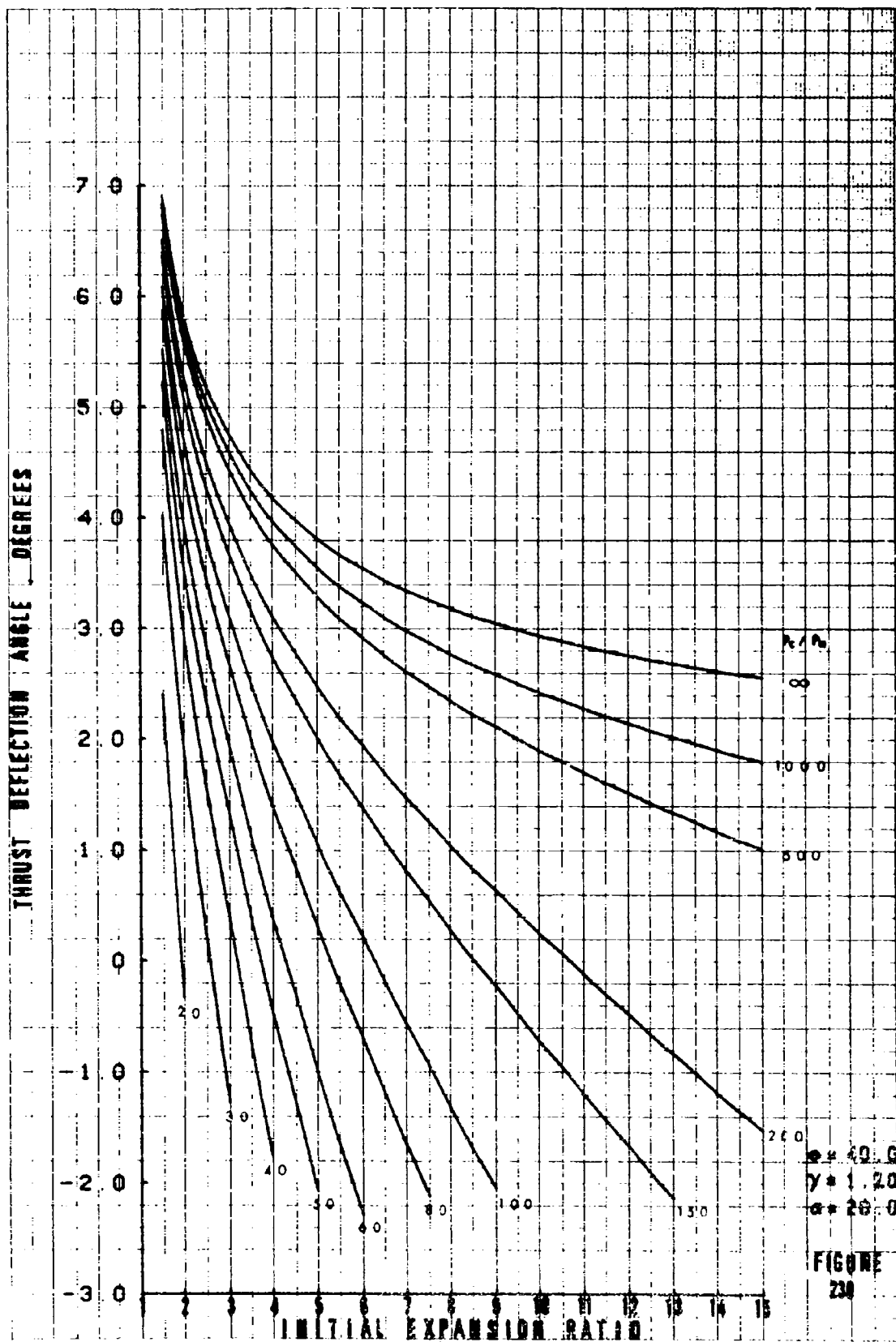


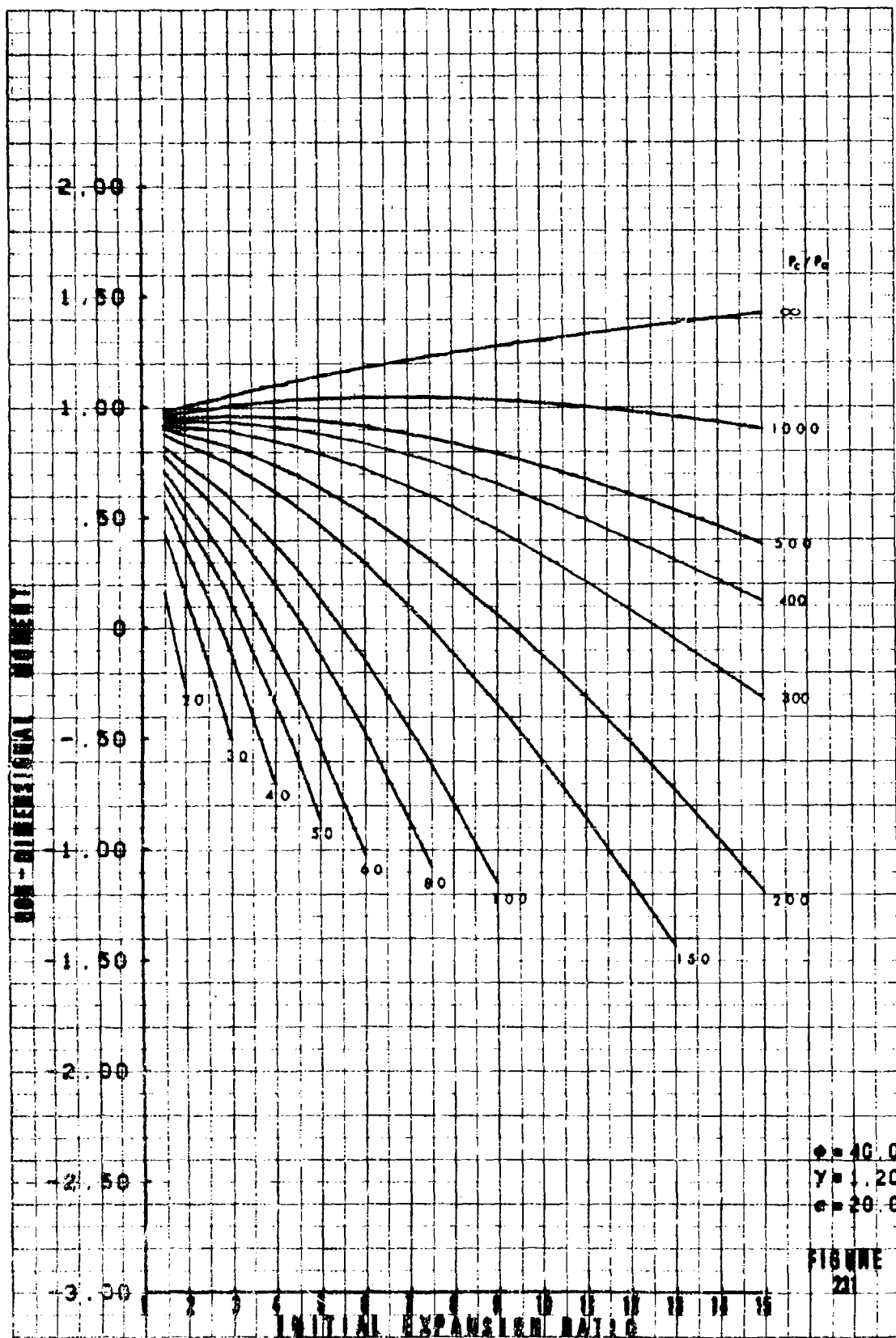
FIGURE 228

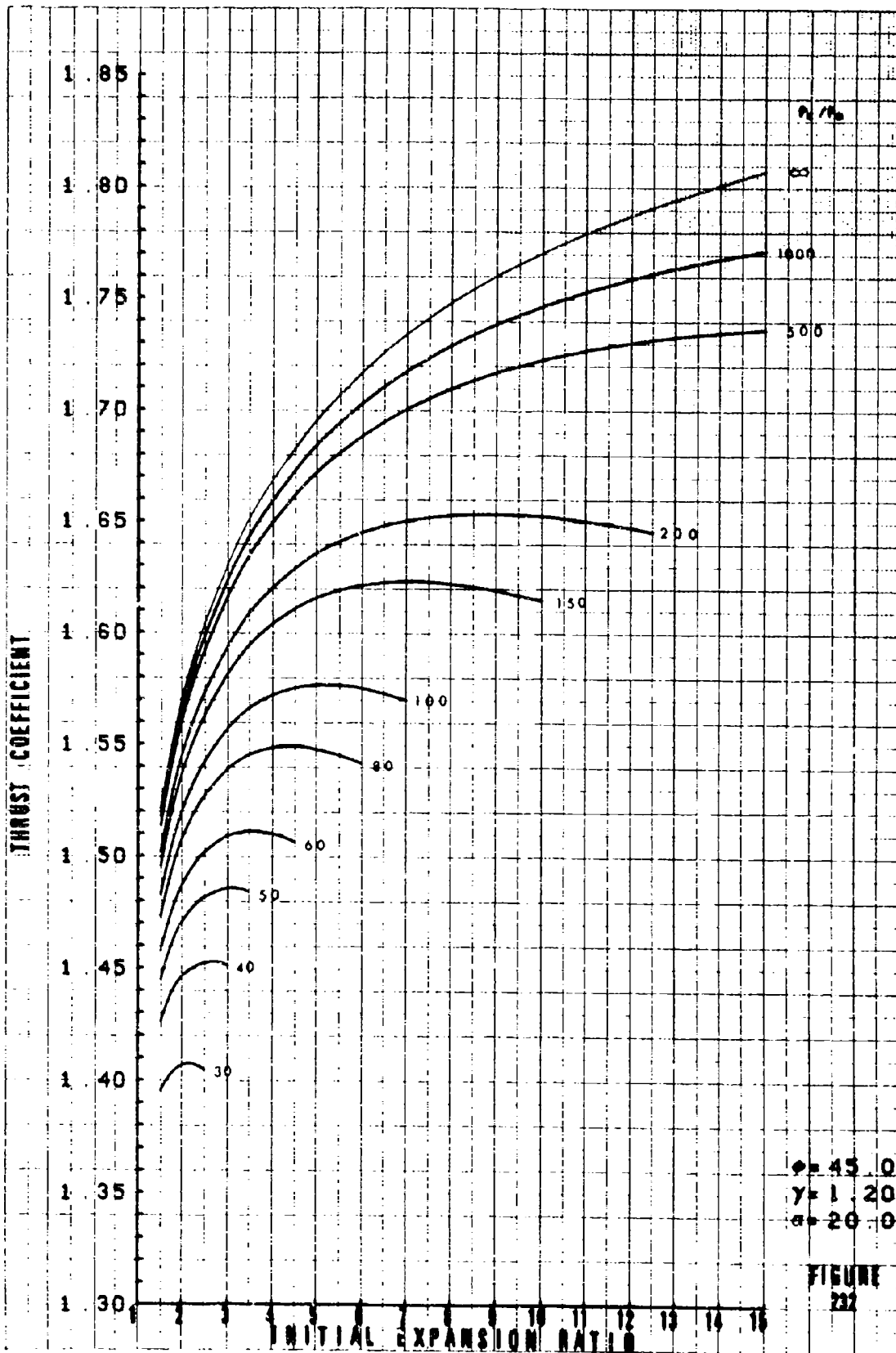


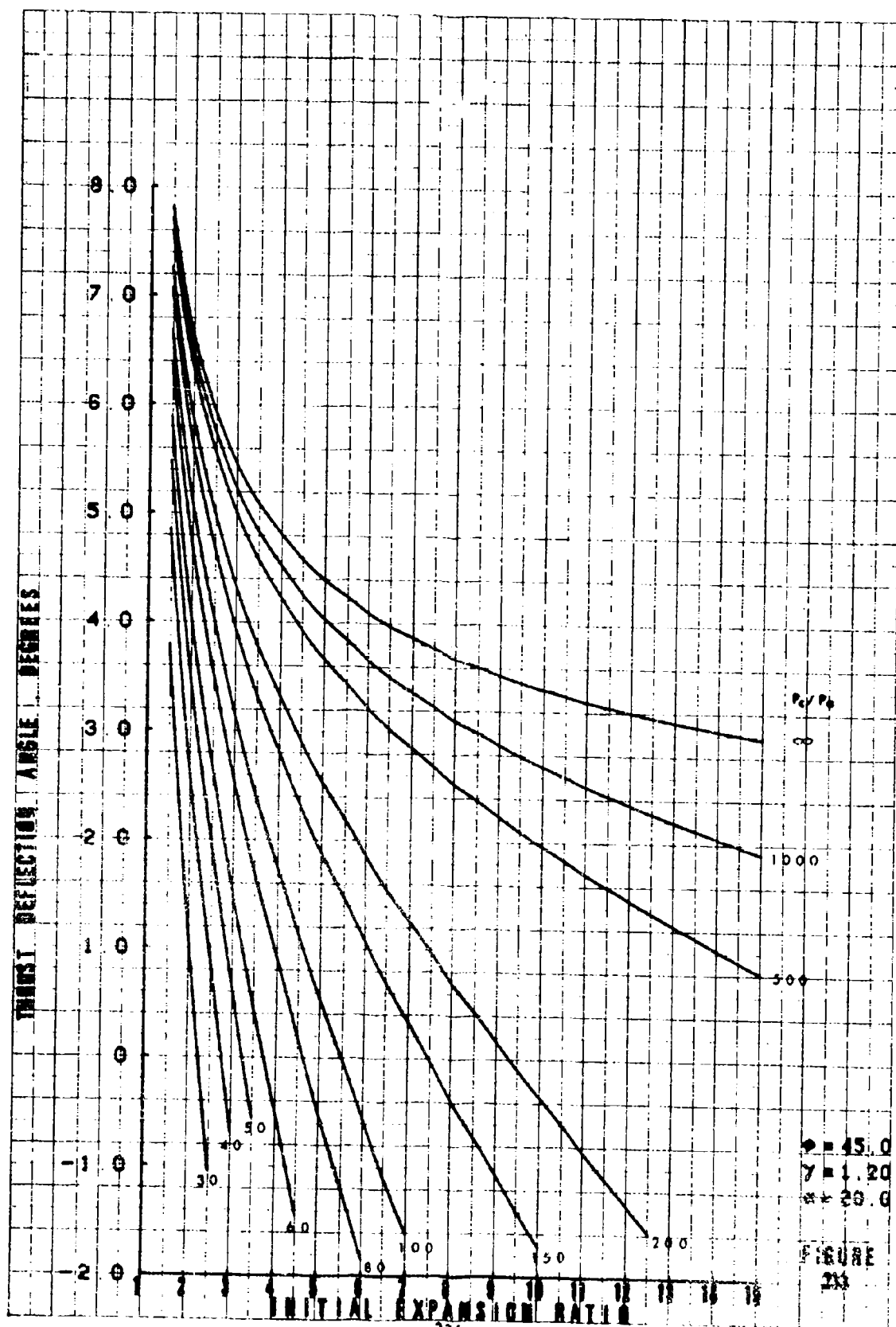


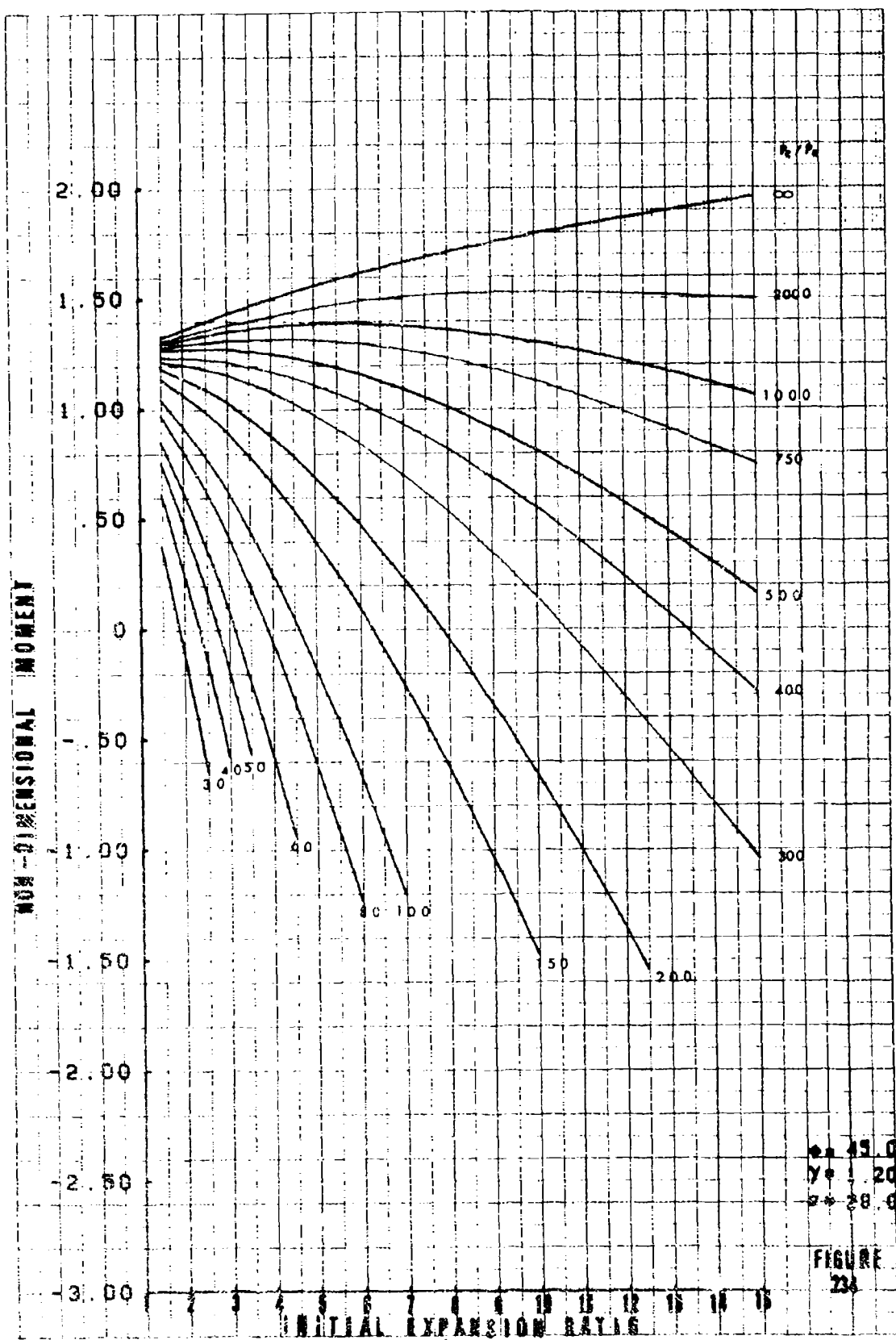






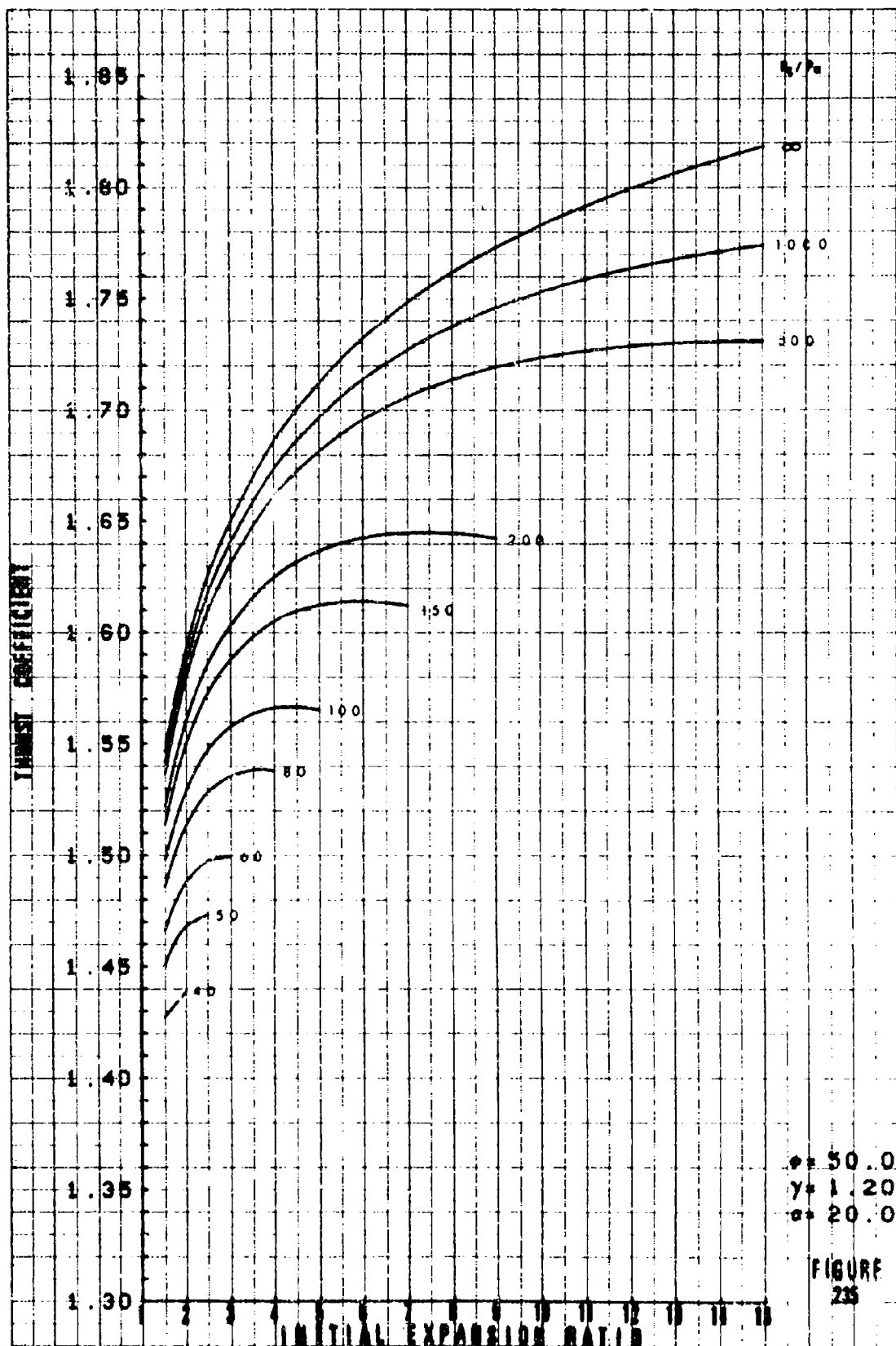


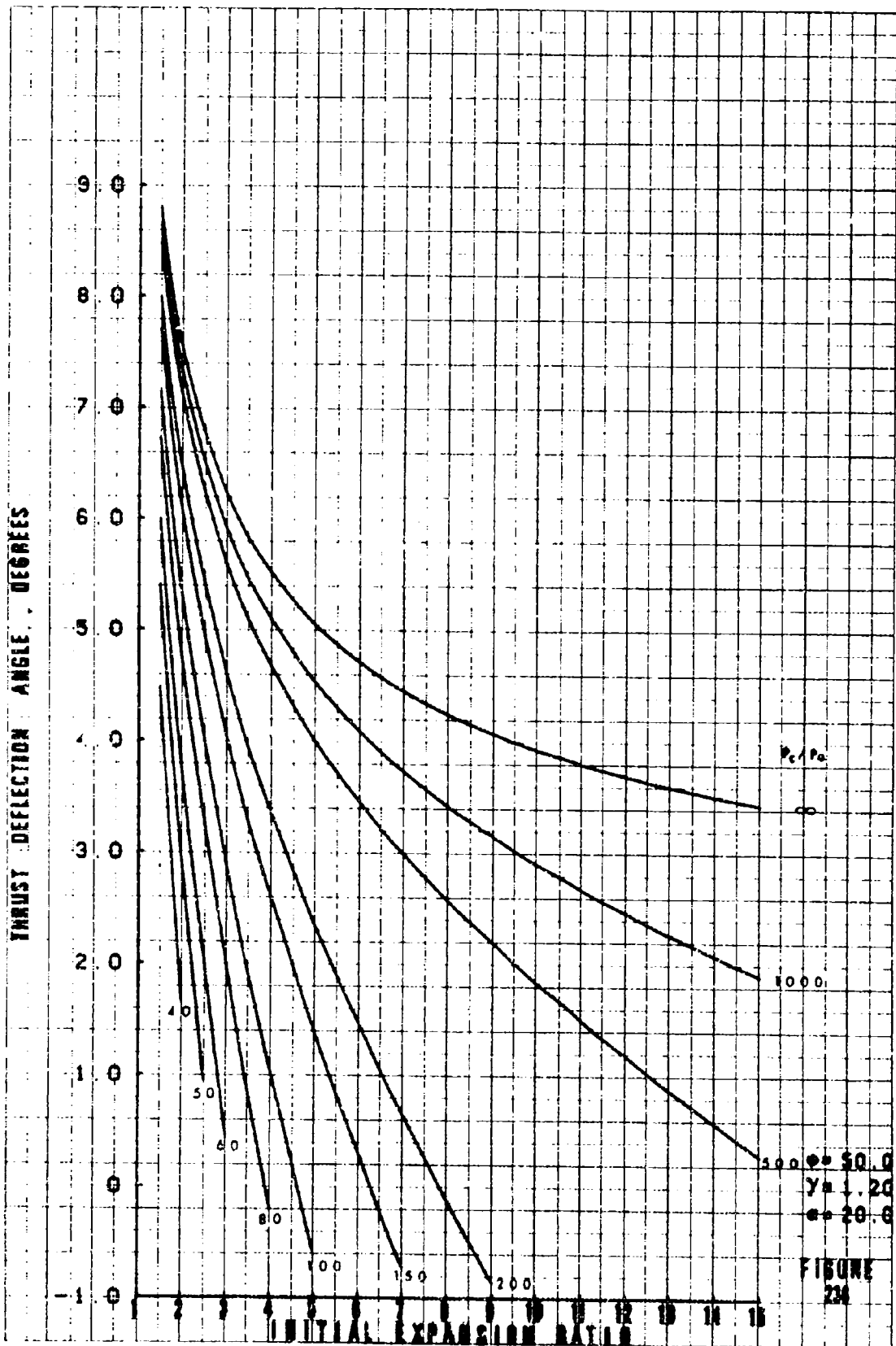


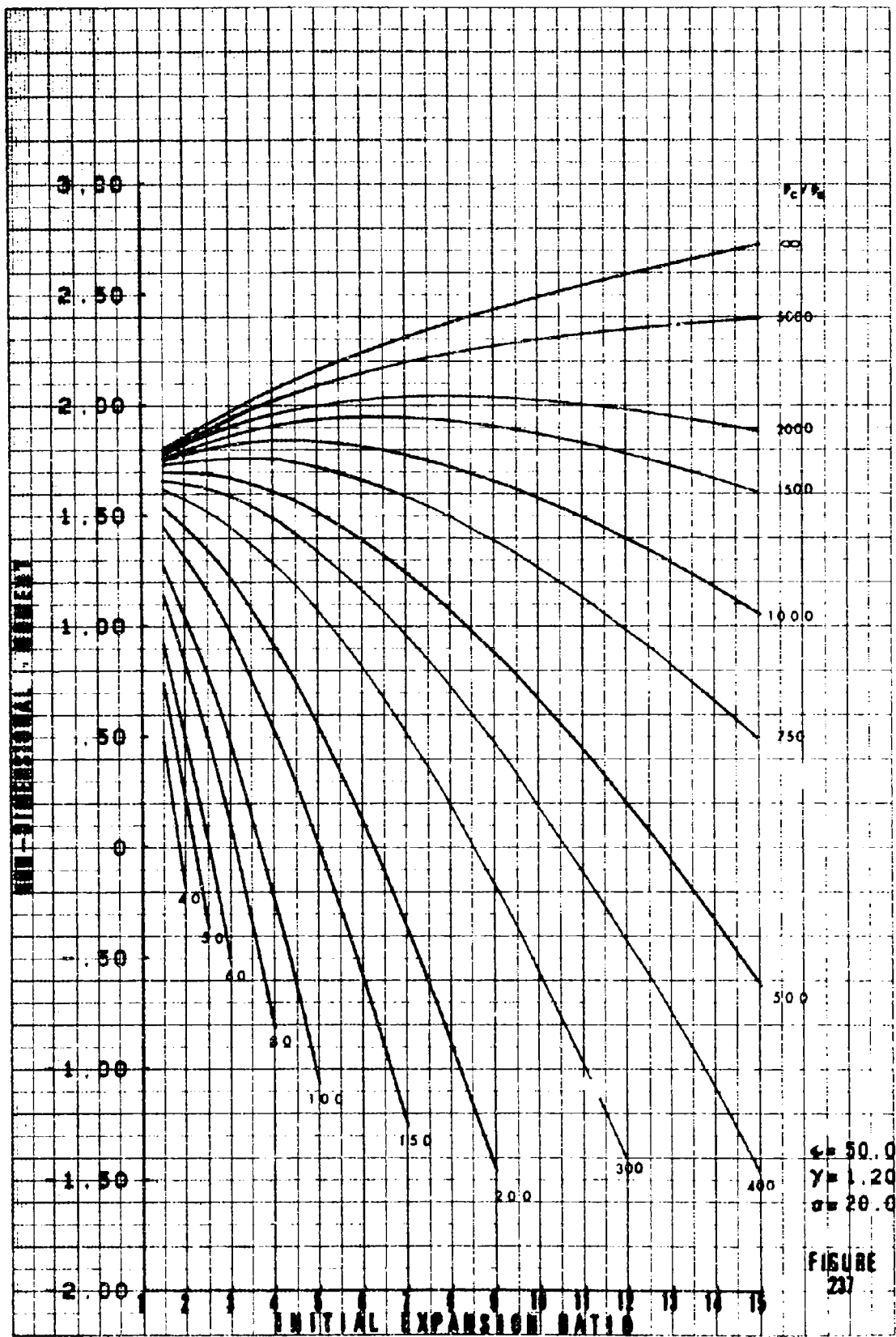


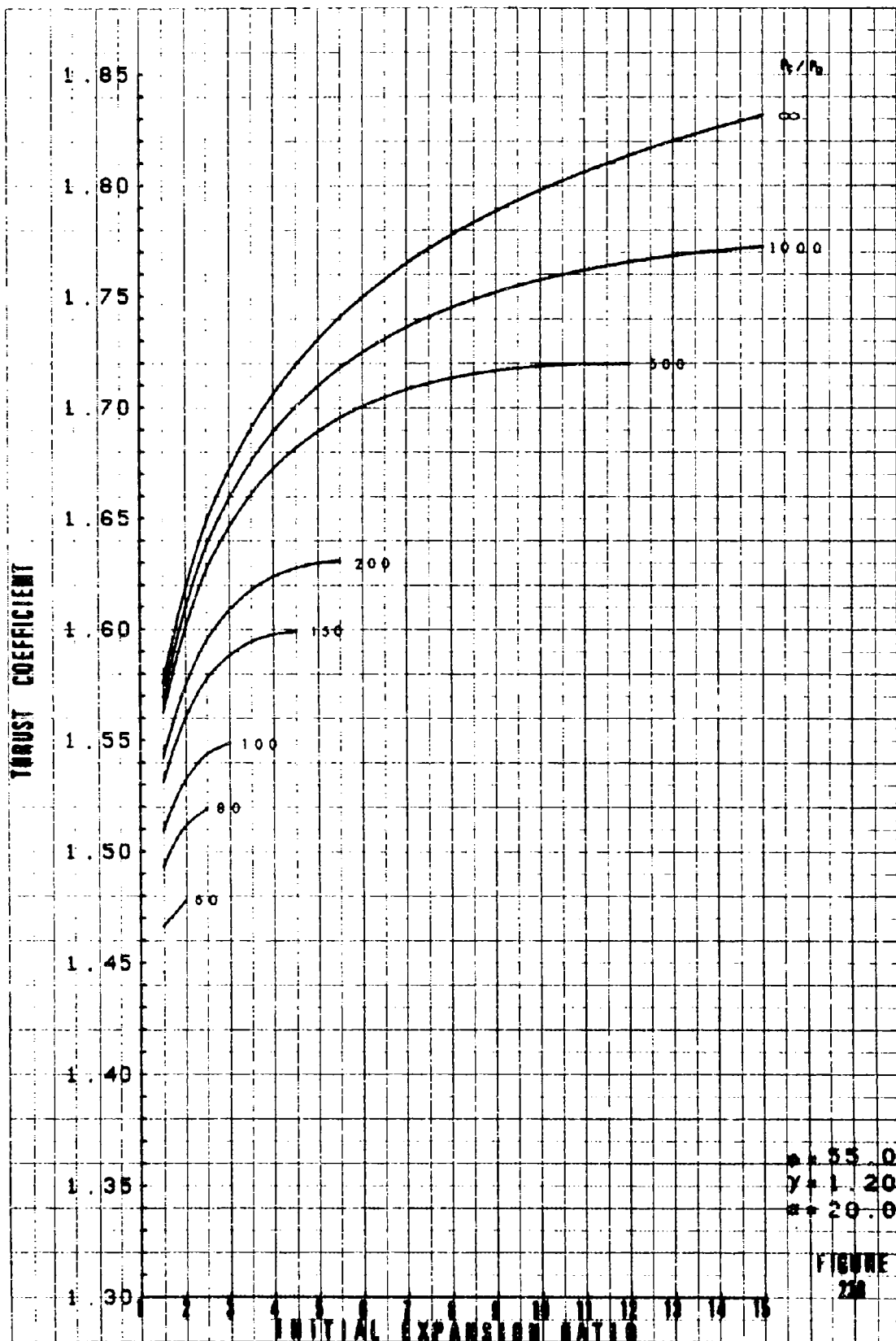
$\alpha = 45.0$
 $\gamma = 1.20$
 $\beta = 28.0$

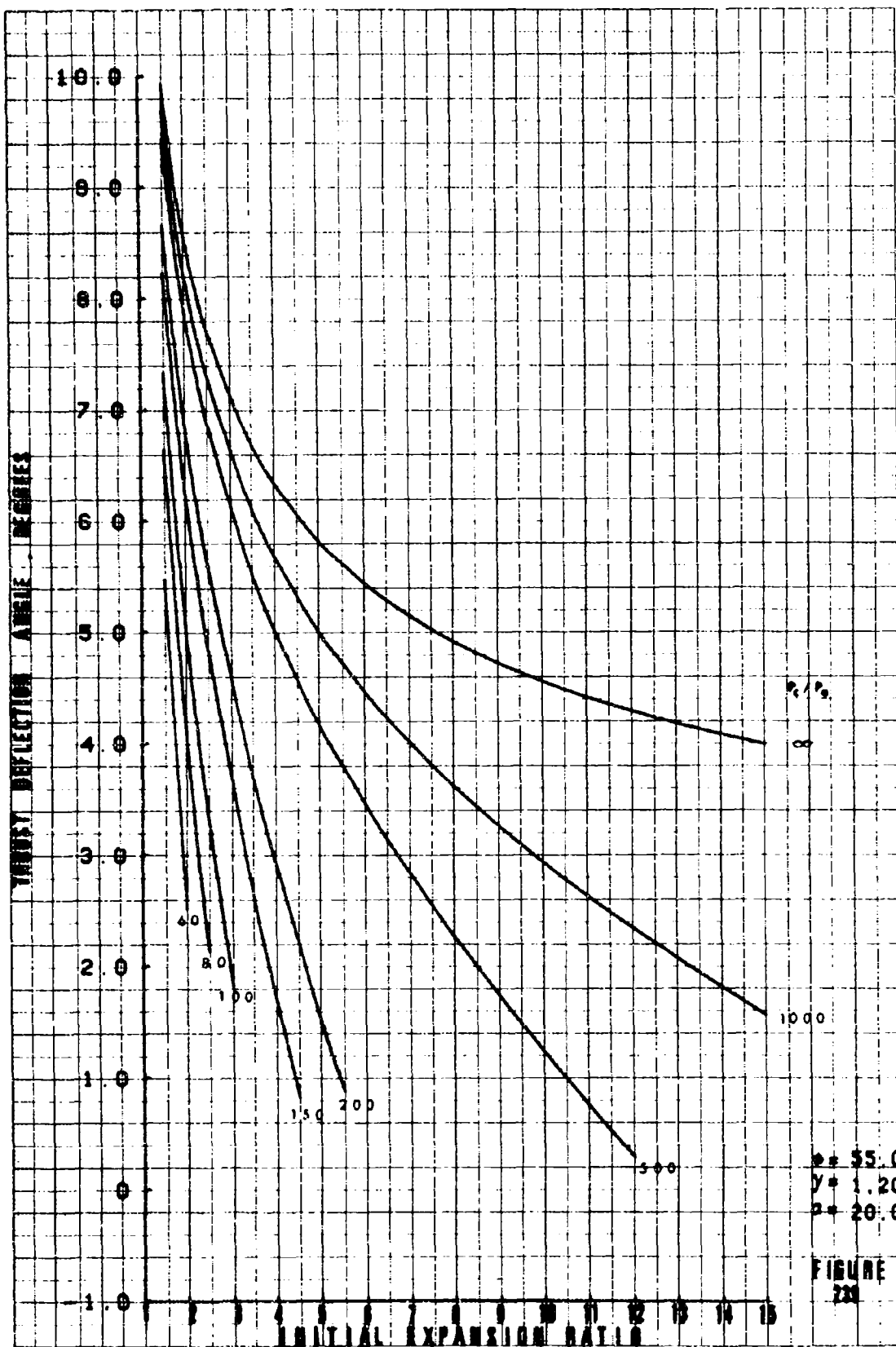
FIGURE 23a

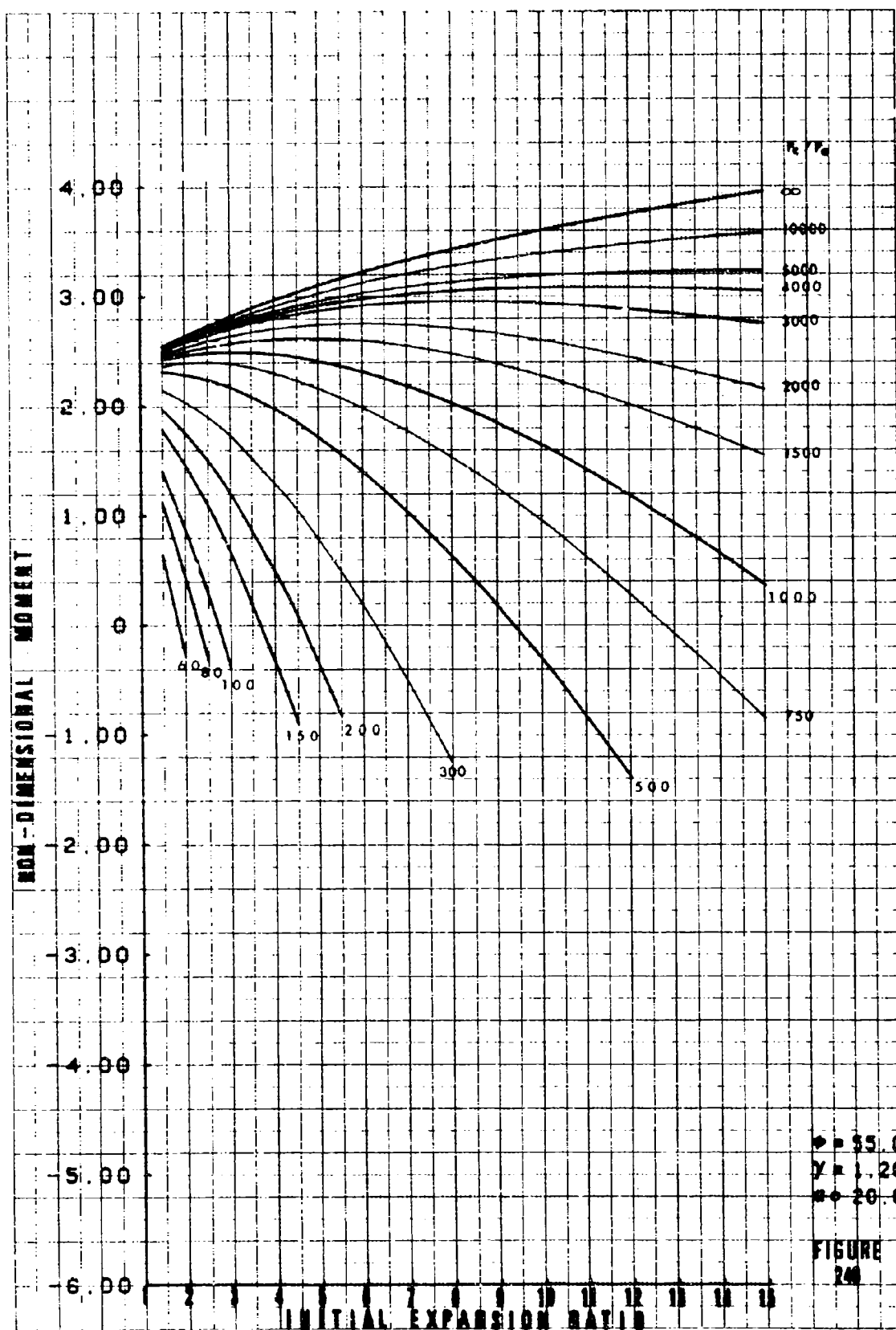


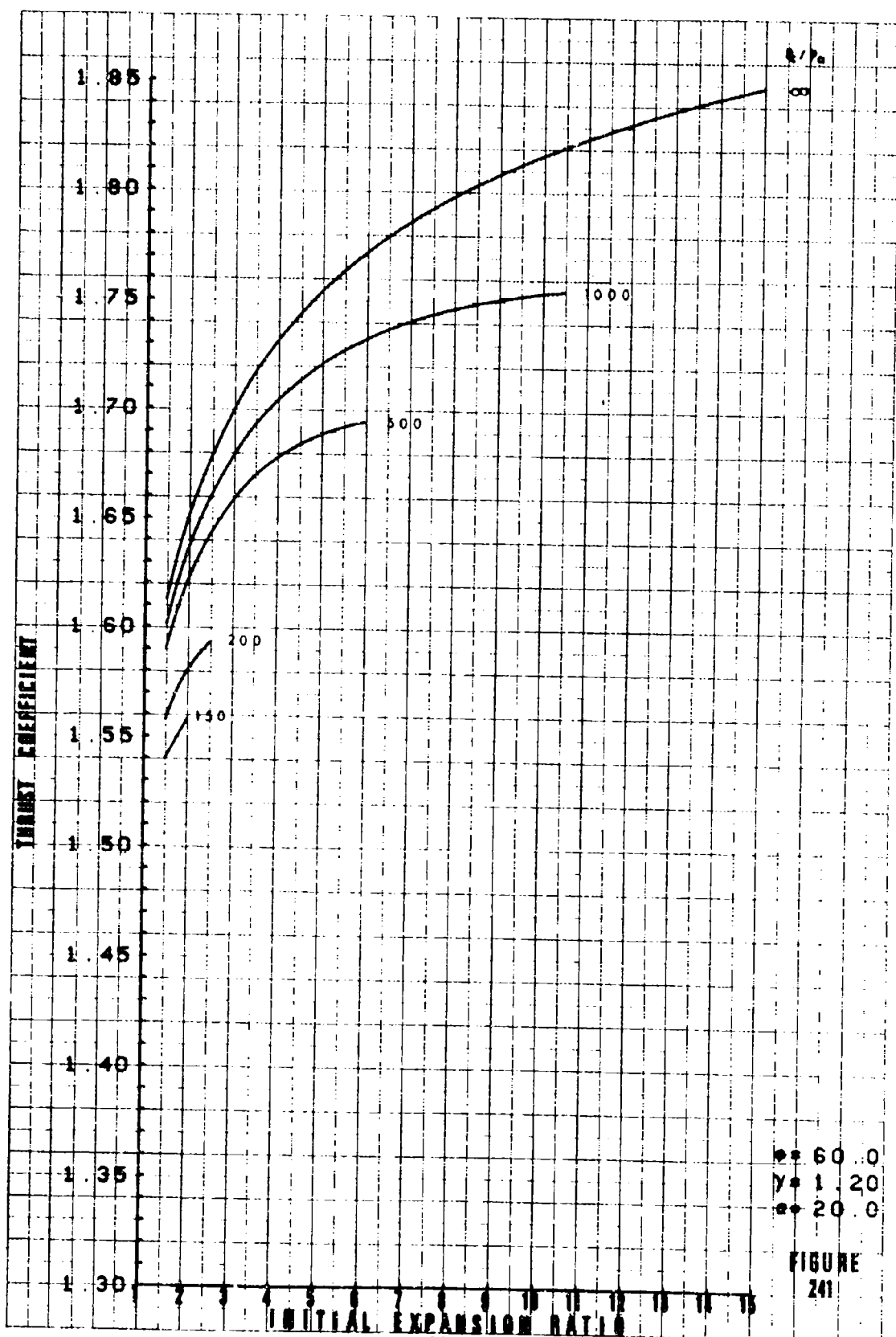






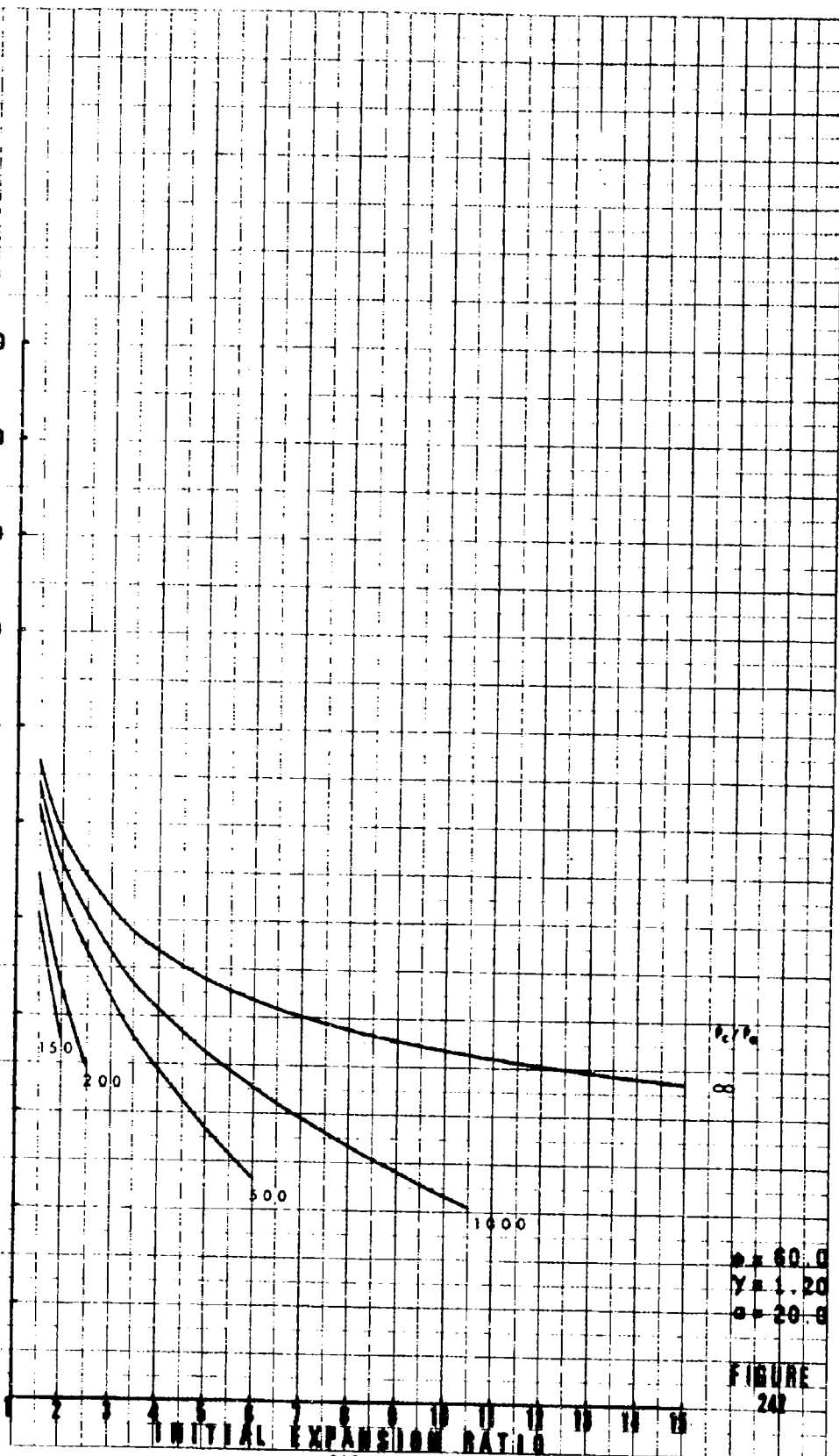






THRUST DEFLECTION ANGLE, DEGREES

20.0
18.0
16.0
14.0
12.0
10.0
8.0
6.0
4.0
2.0
0
-2.0

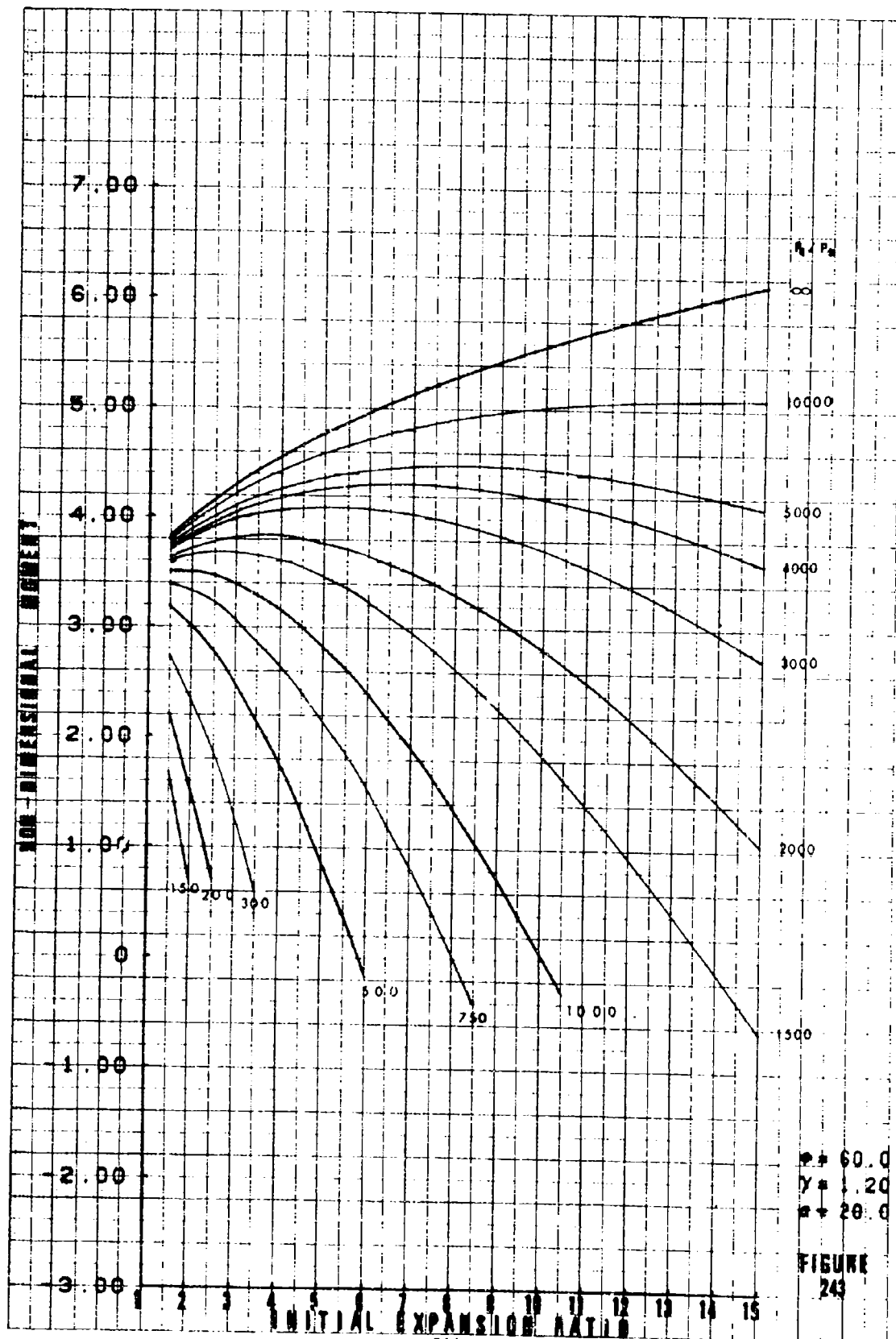


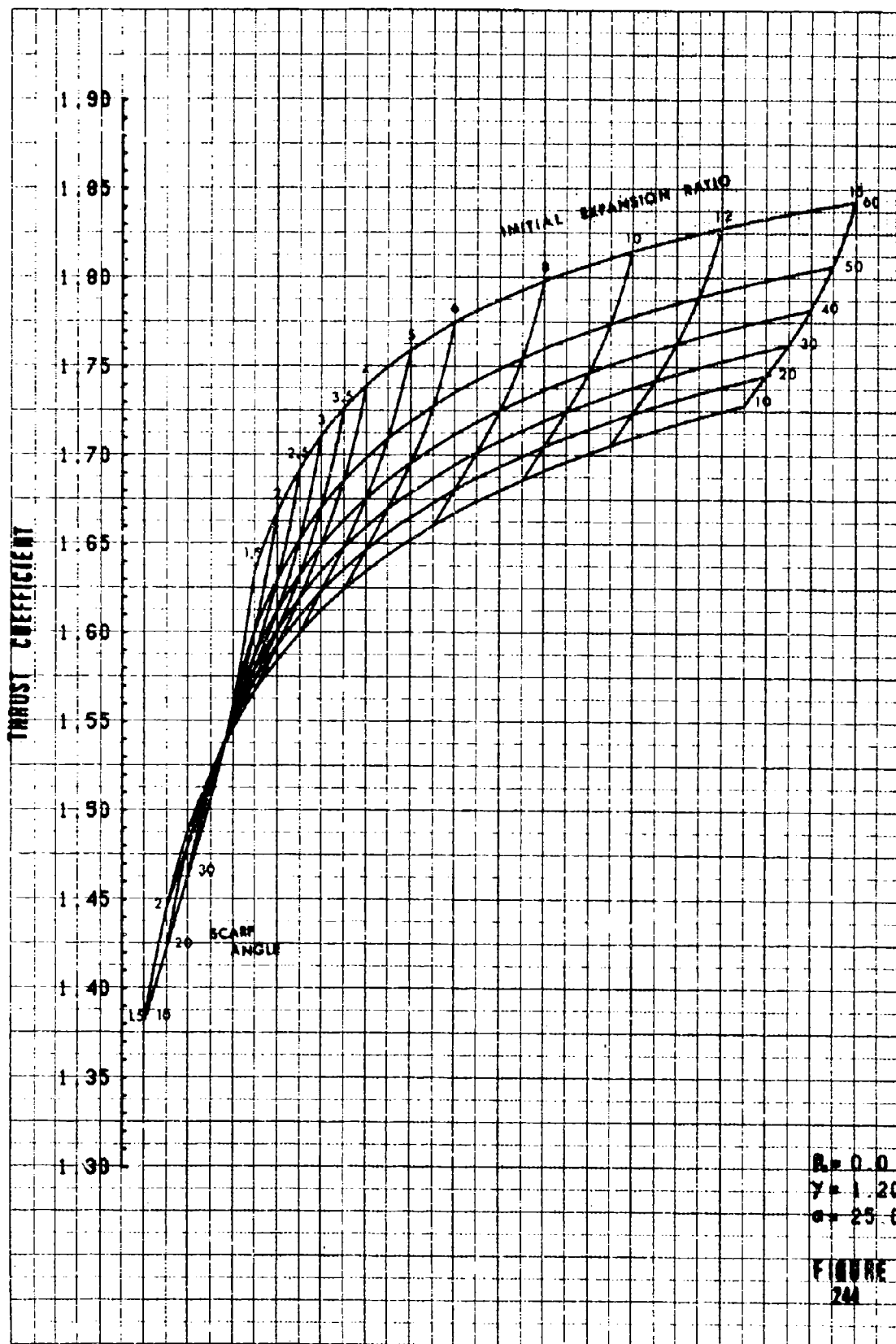
P_c/P_a
8

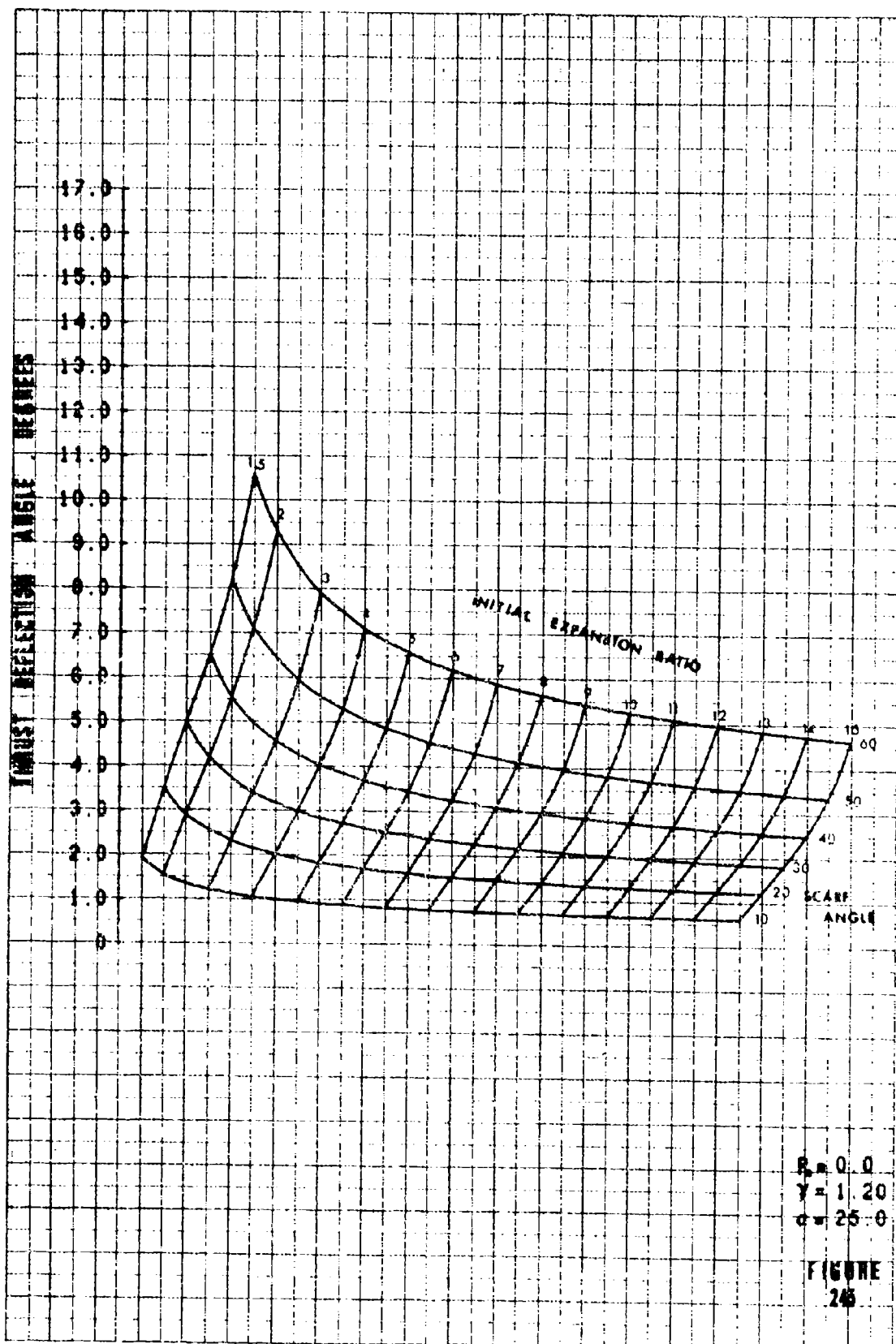
$\gamma = 60.0$
 $\gamma = 1.20$
 $\theta = 20.0$

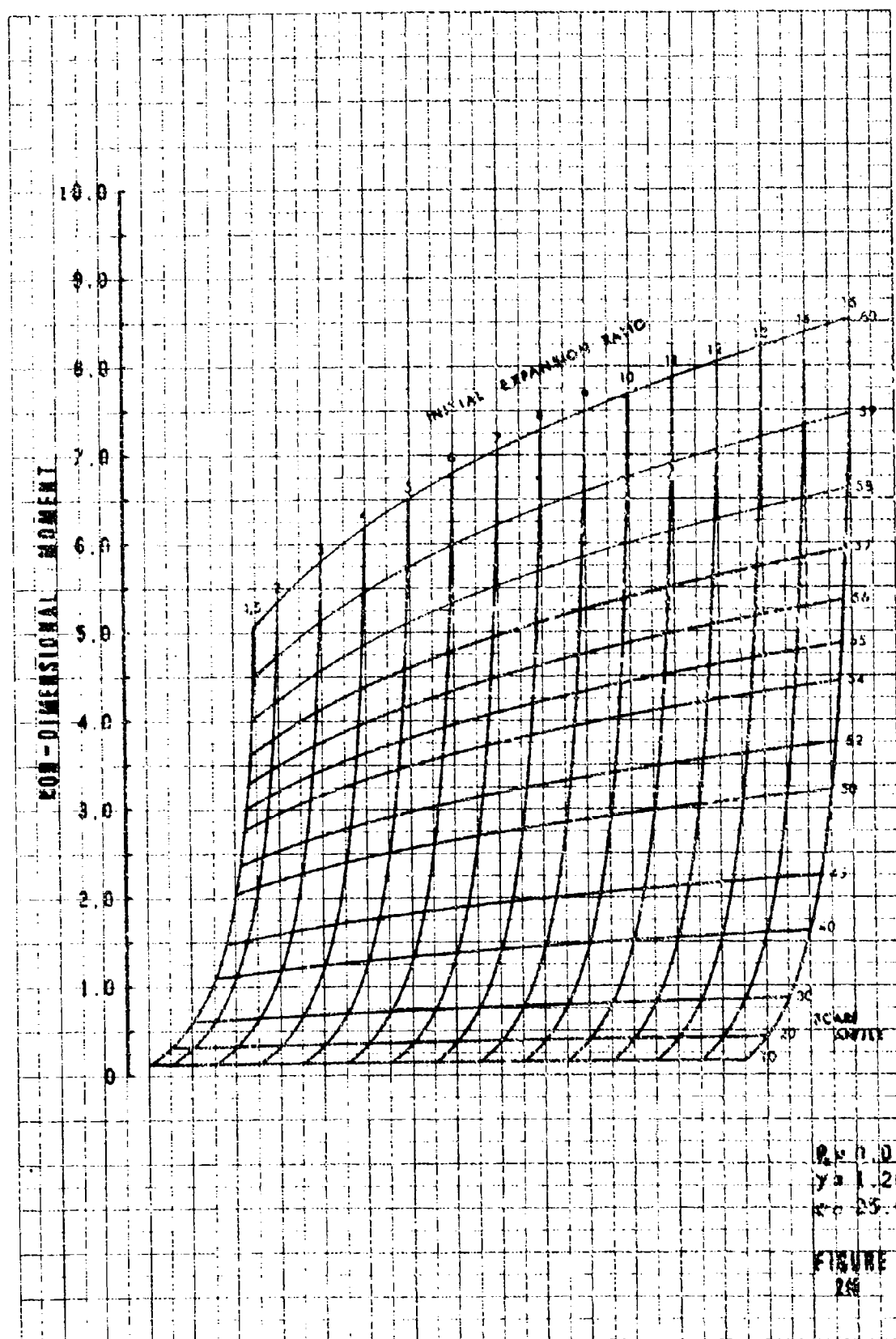
FIGURE
240

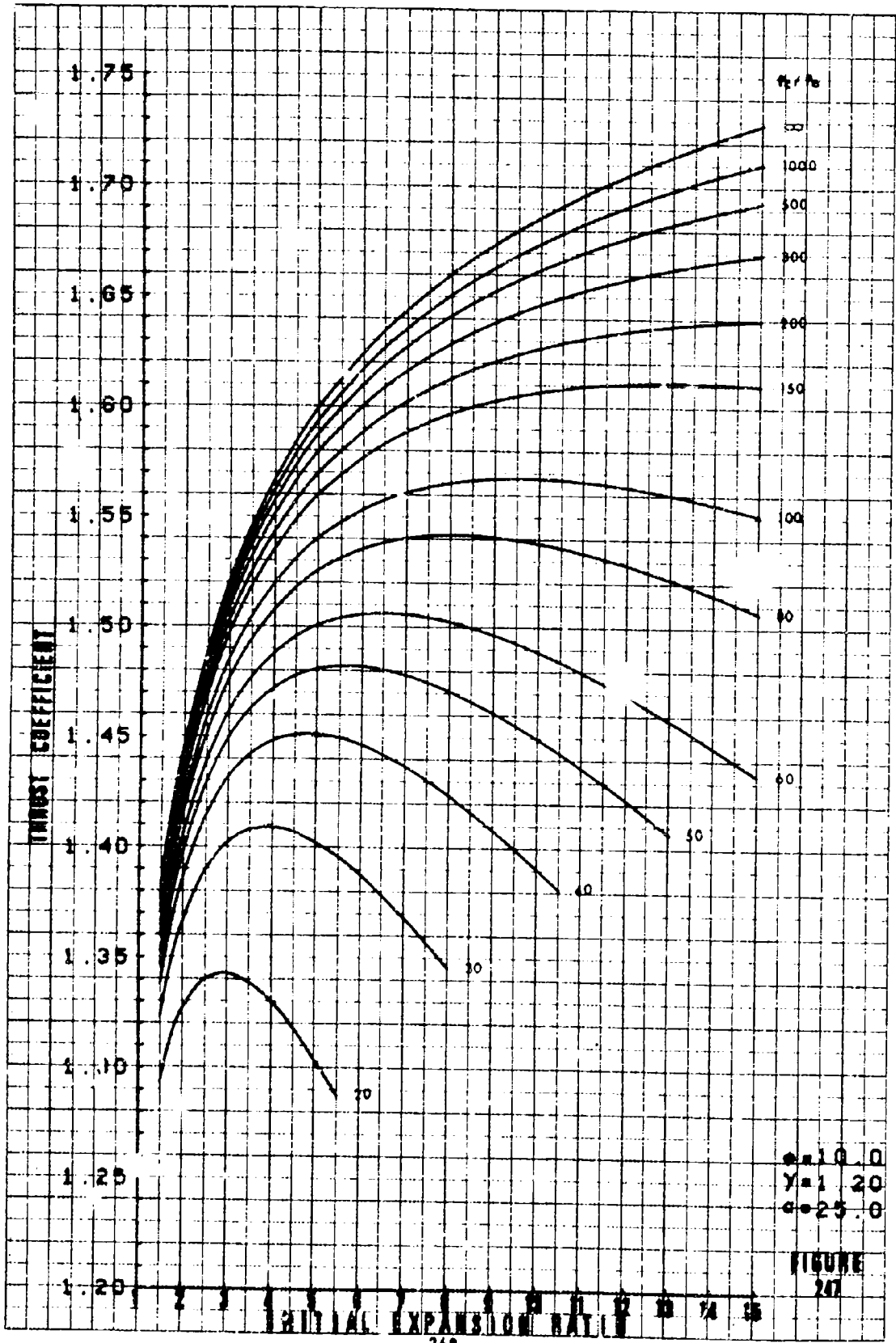
INITIAL EXPANSION RATIO

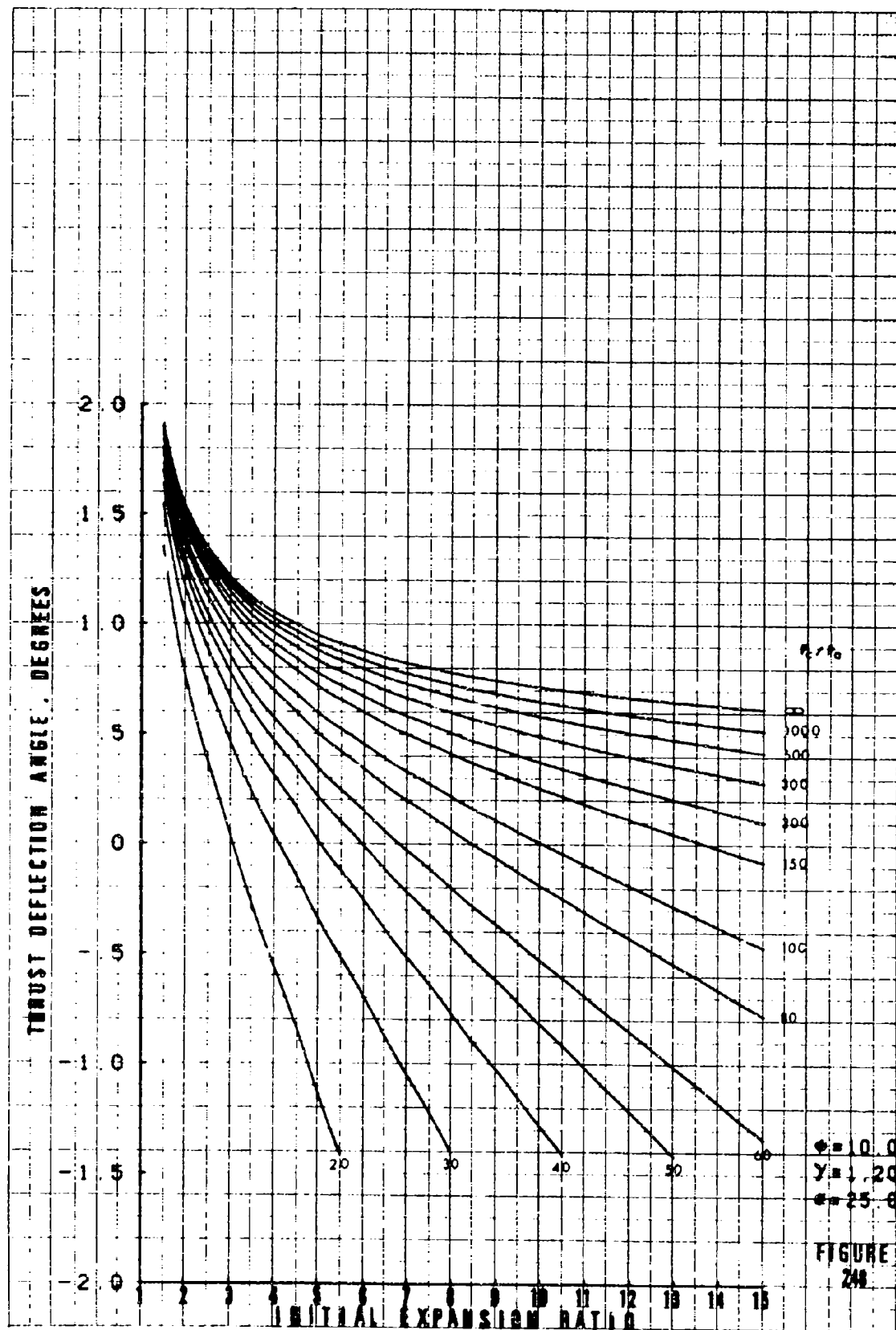


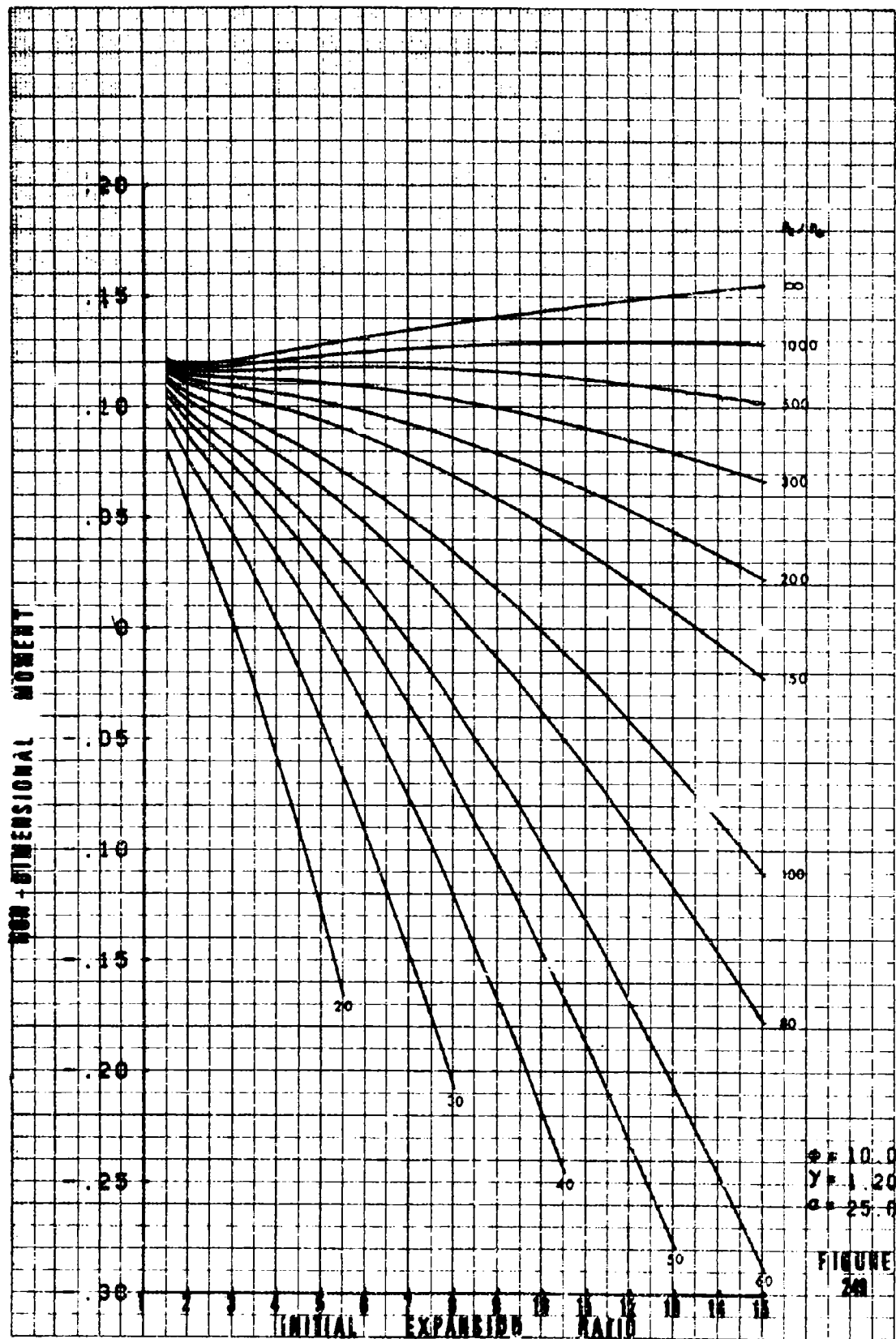






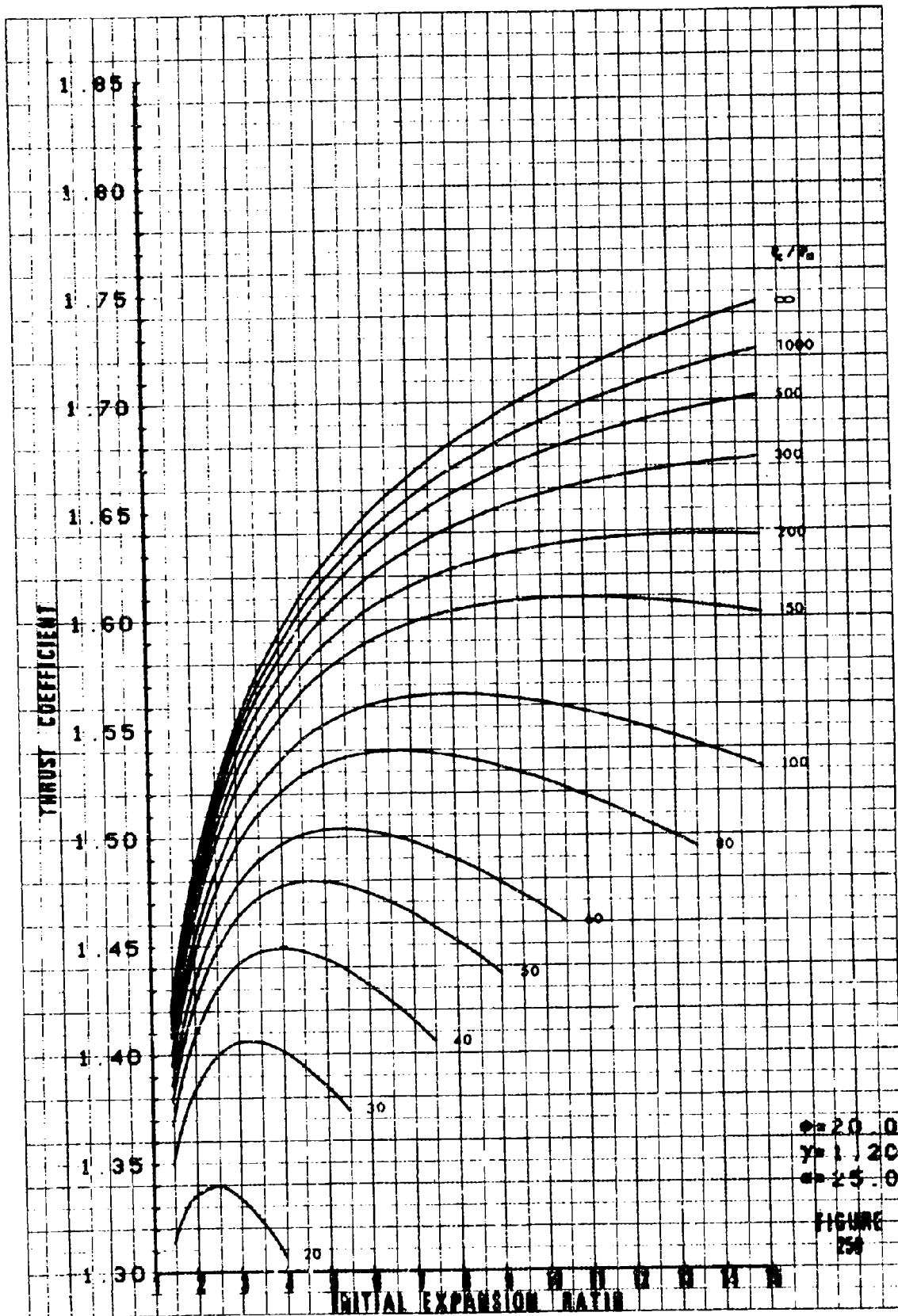






$\phi = 10.0$
 $\gamma = 1.20$
 $\alpha = 25.0$

FIGURE
249



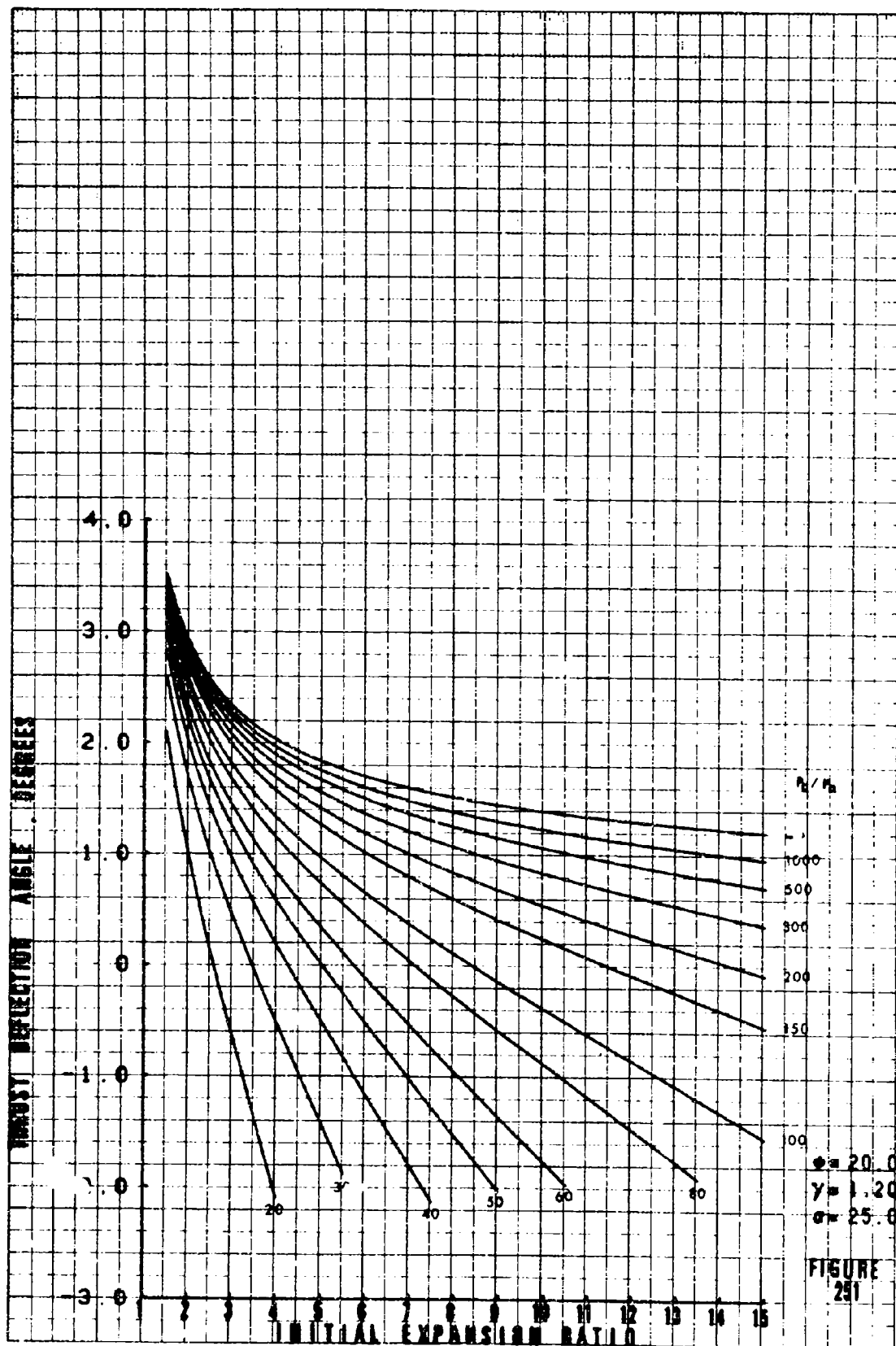
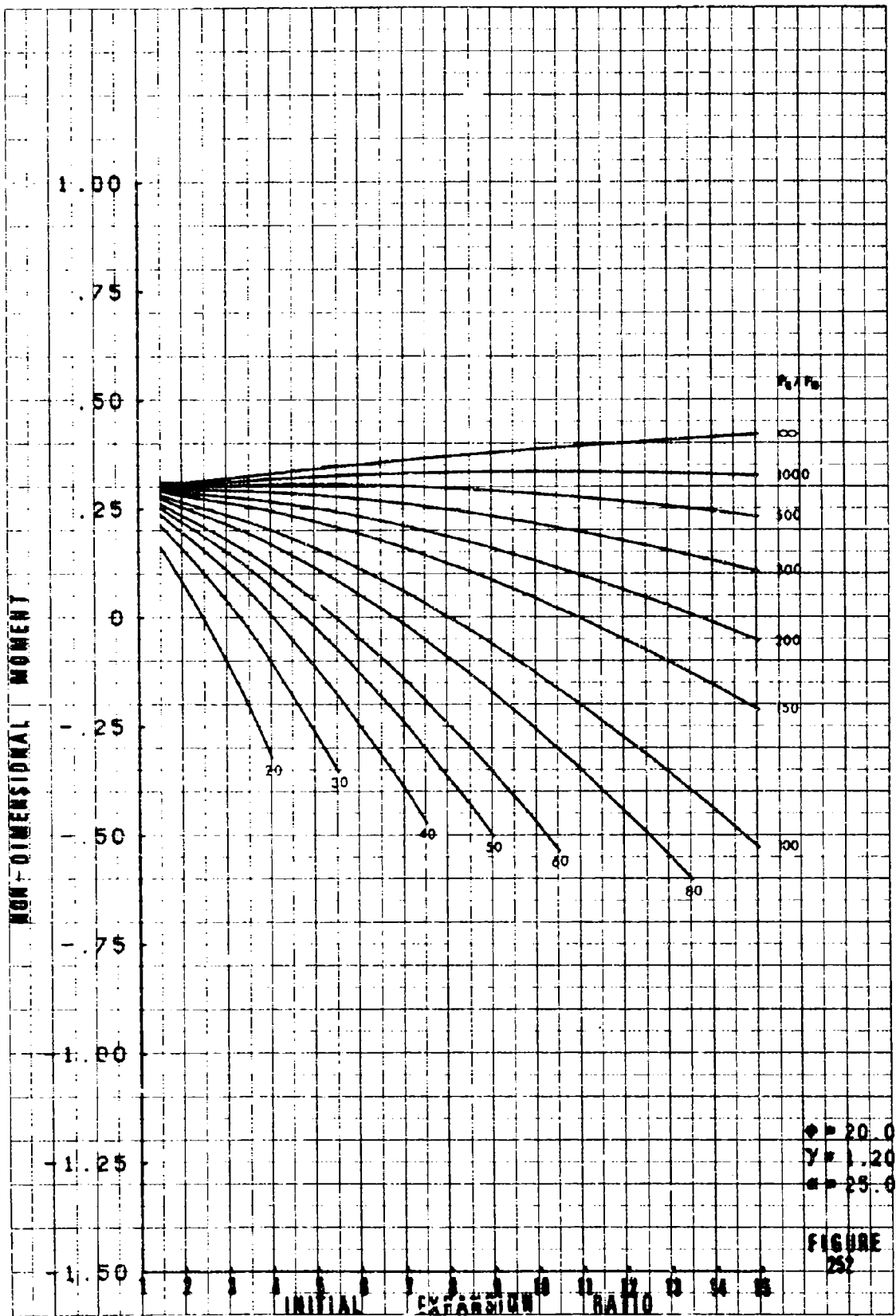
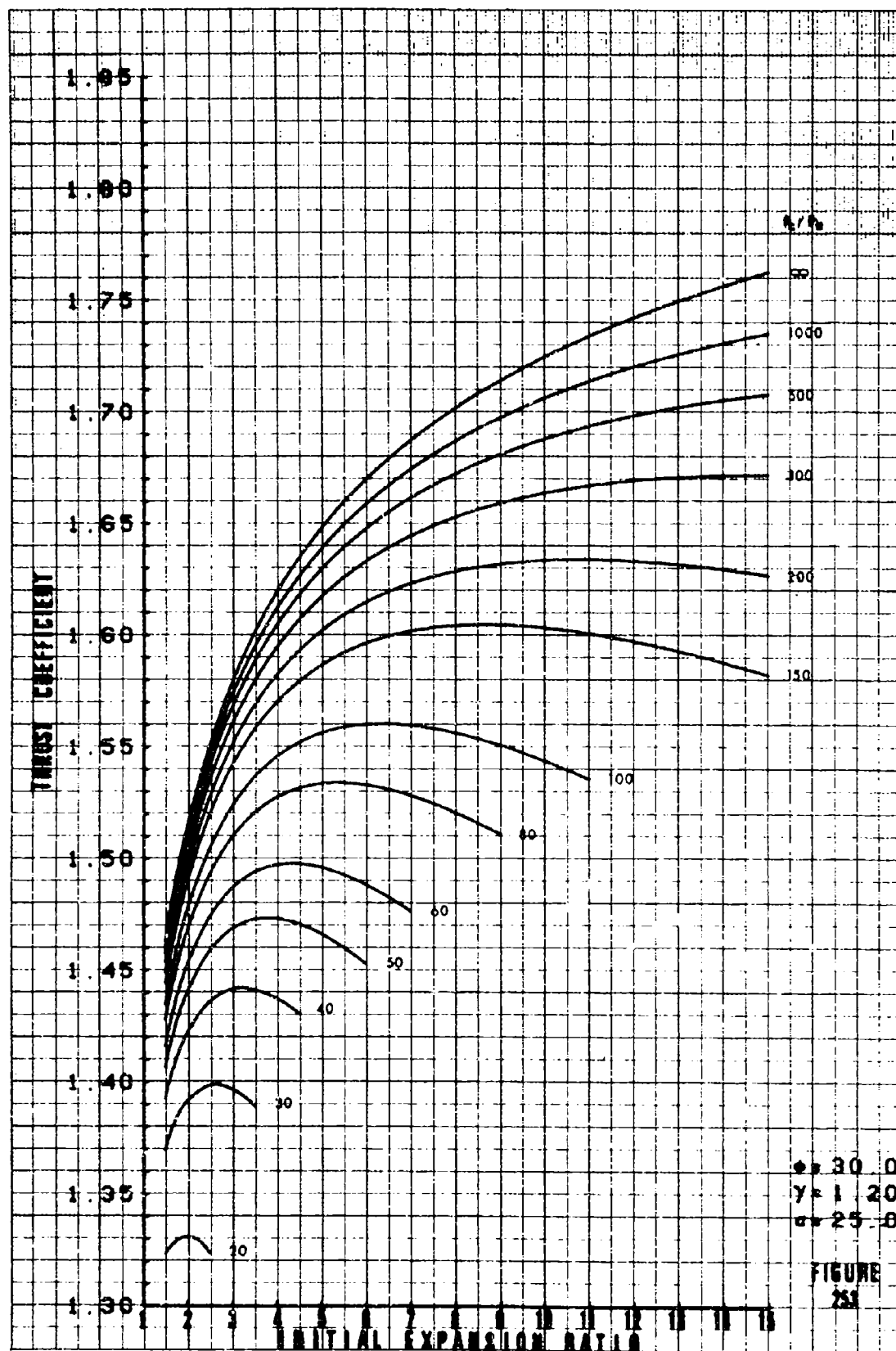
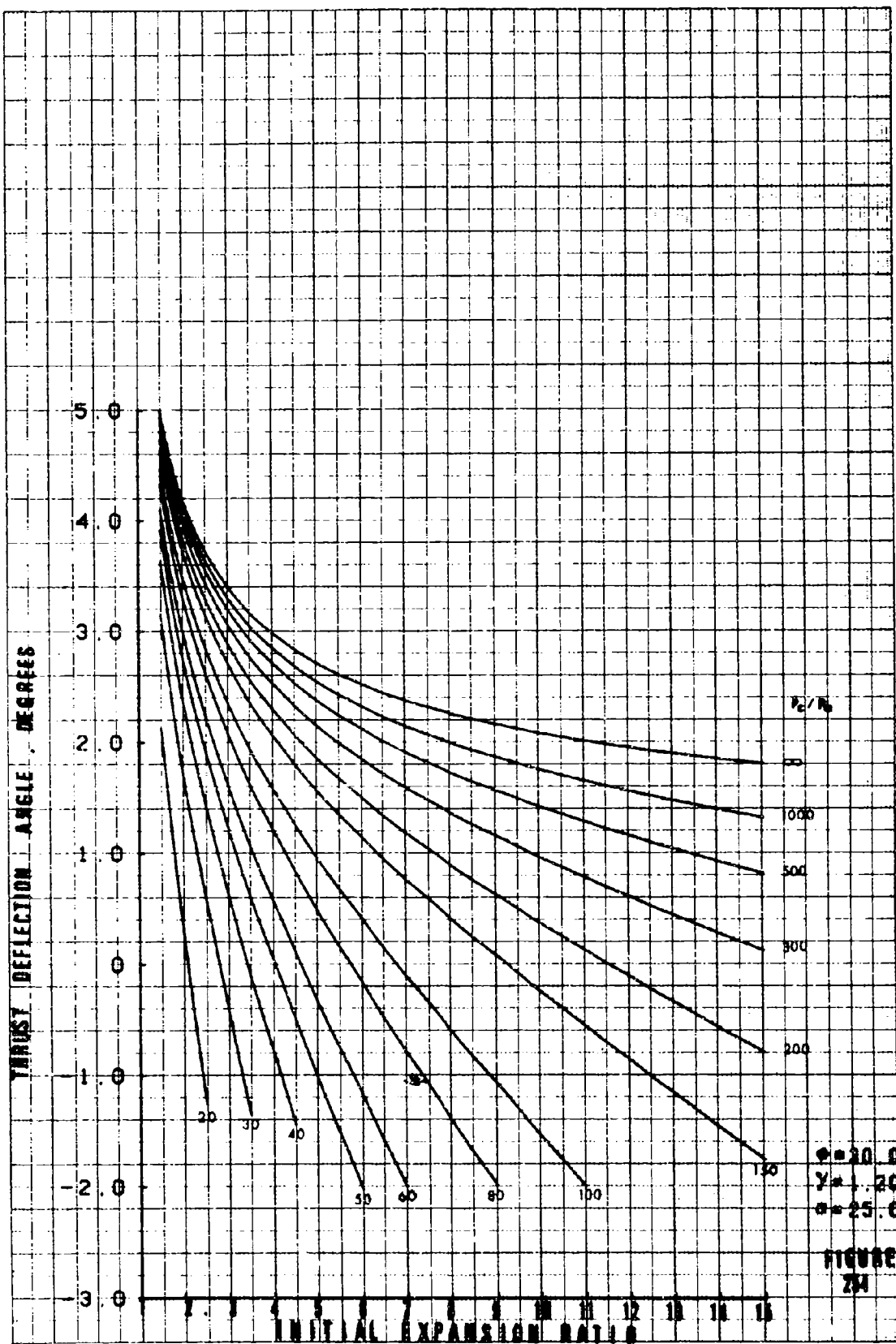


FIGURE 251







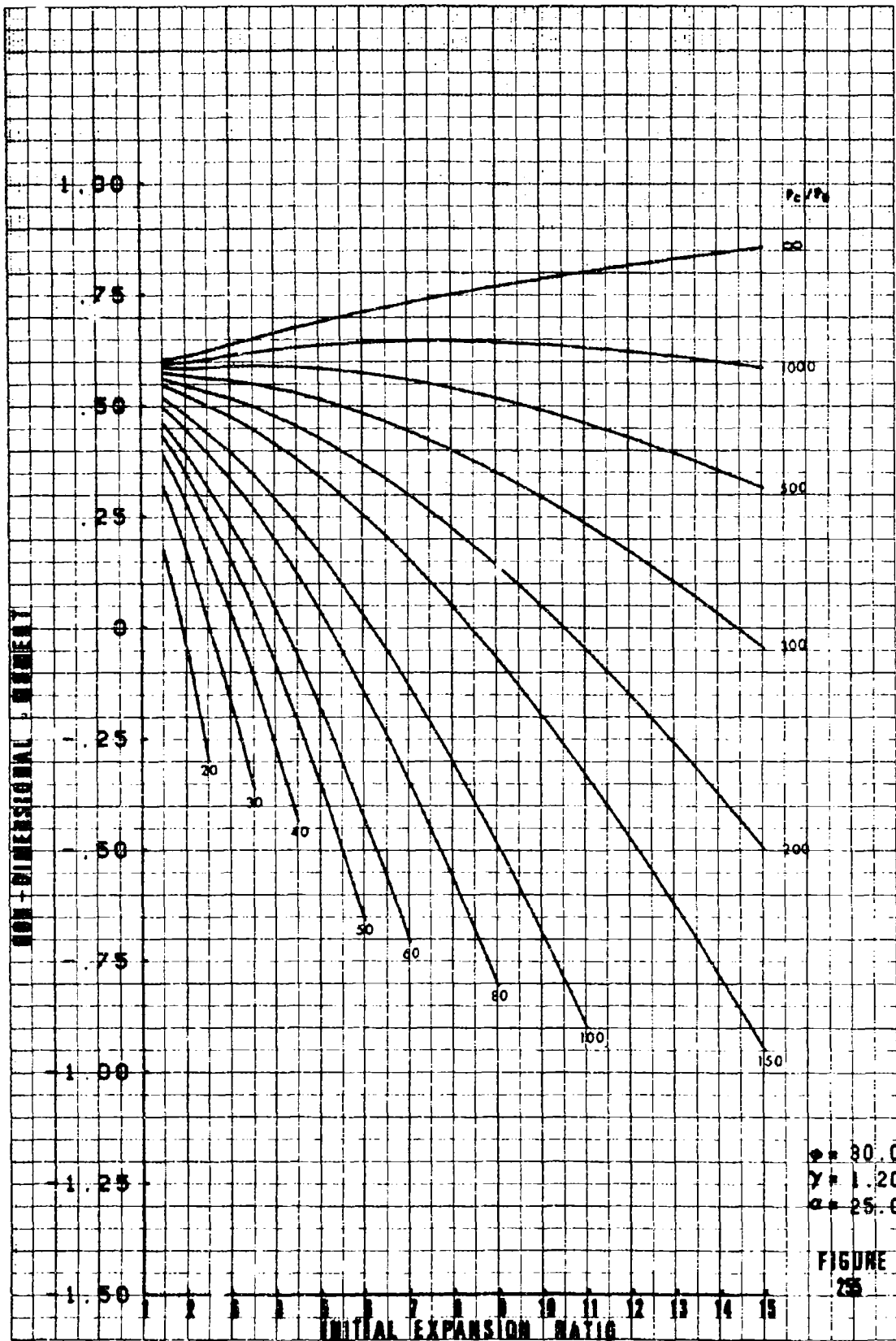
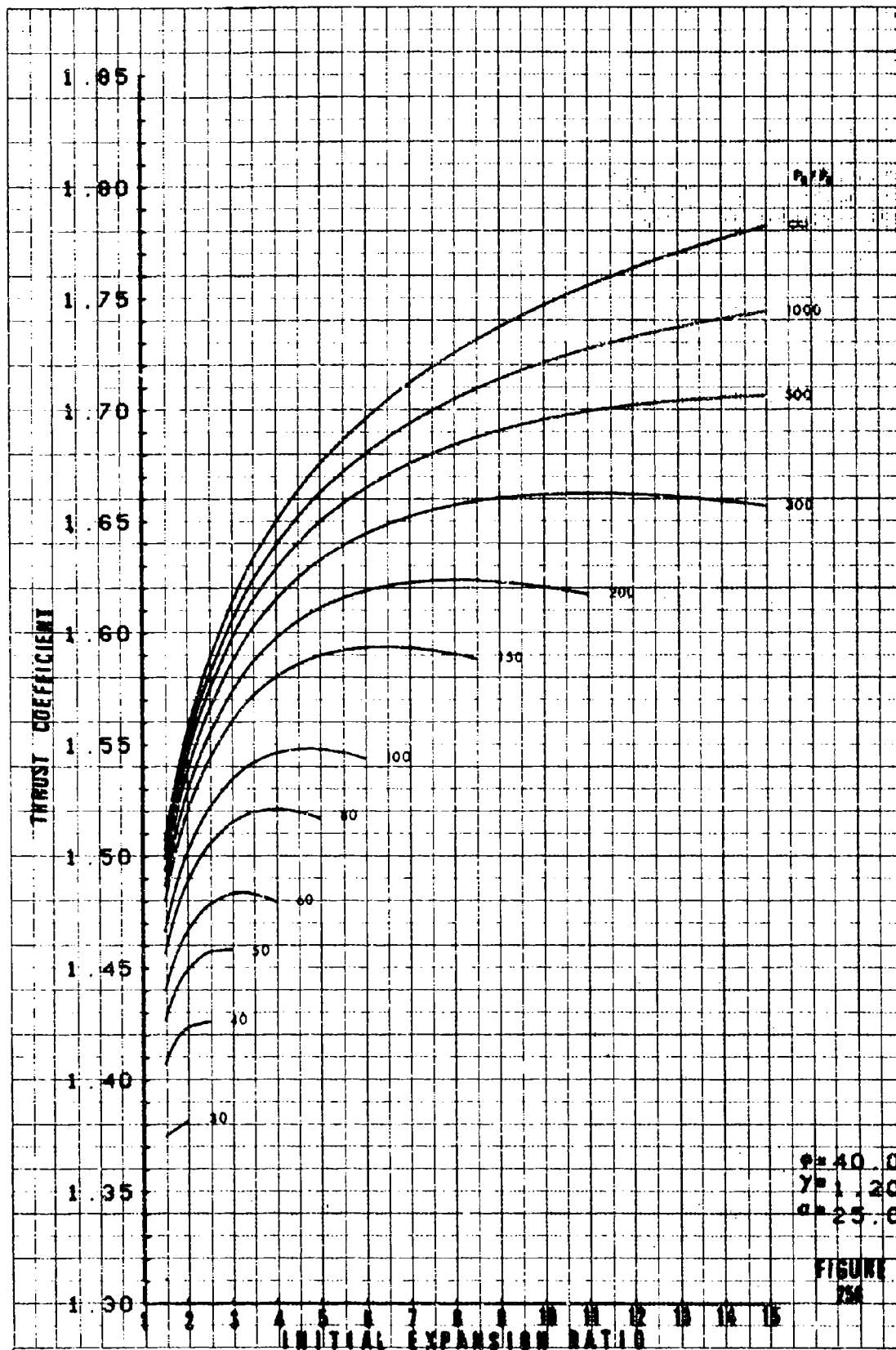
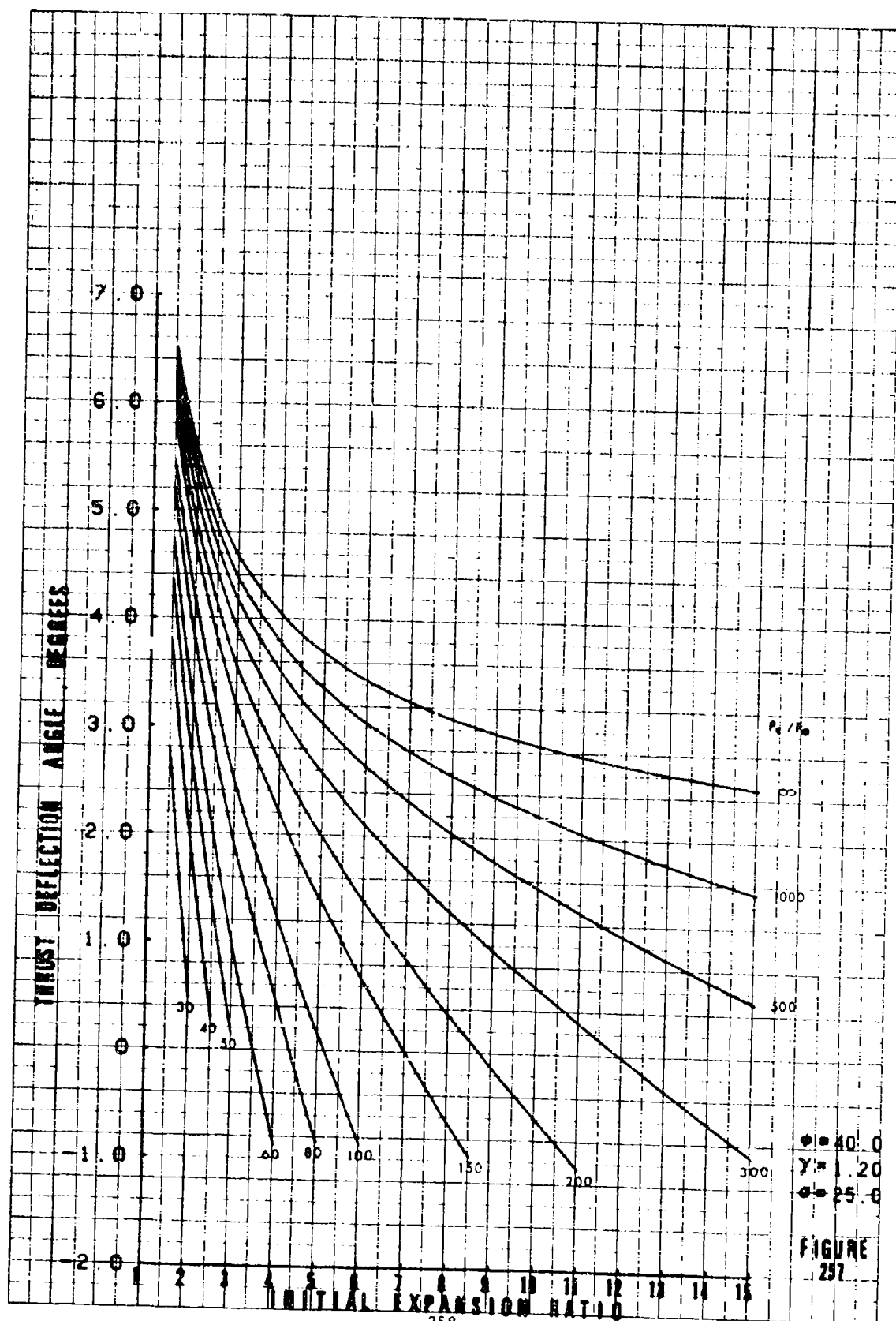
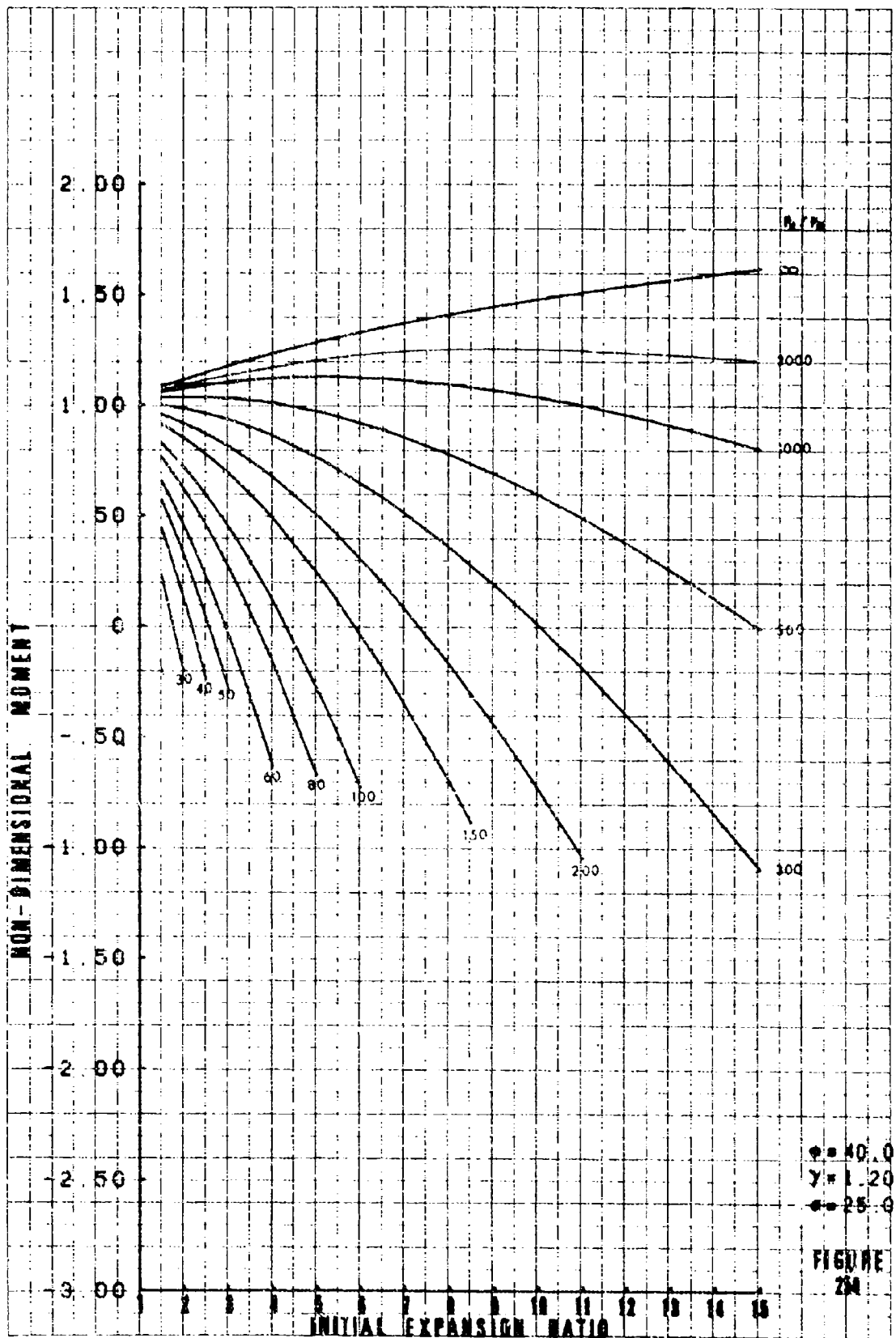
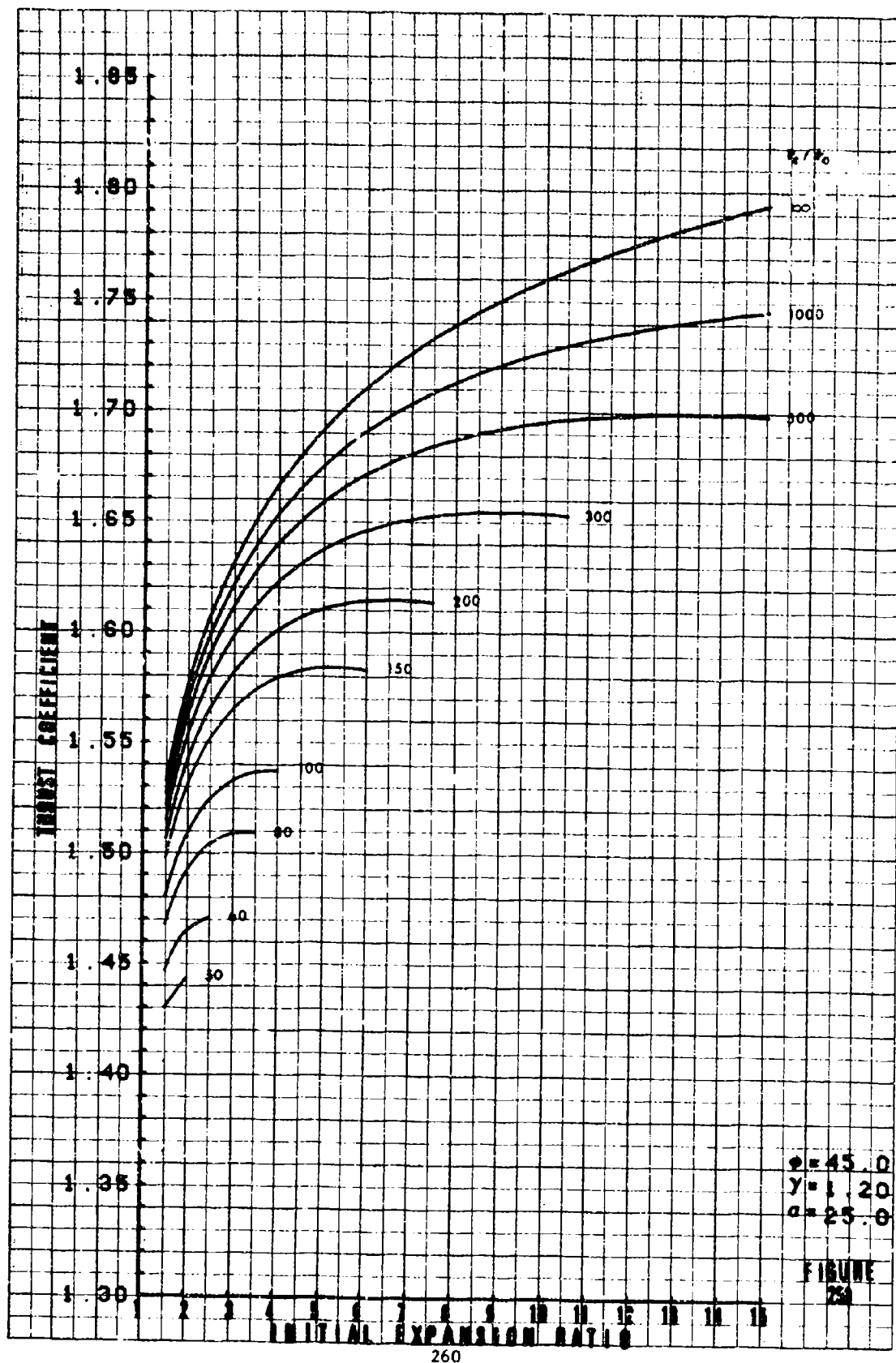


FIGURE 256









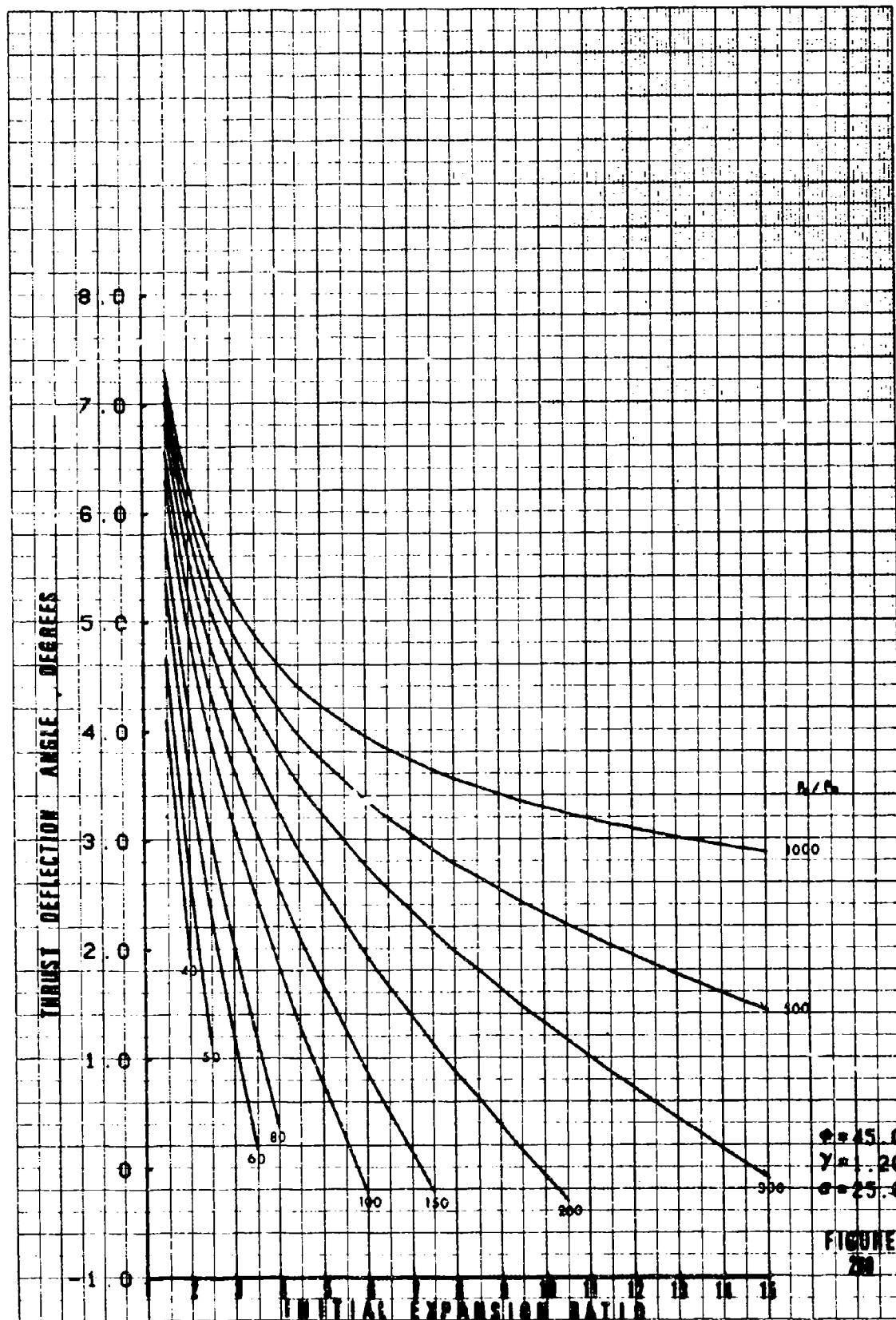
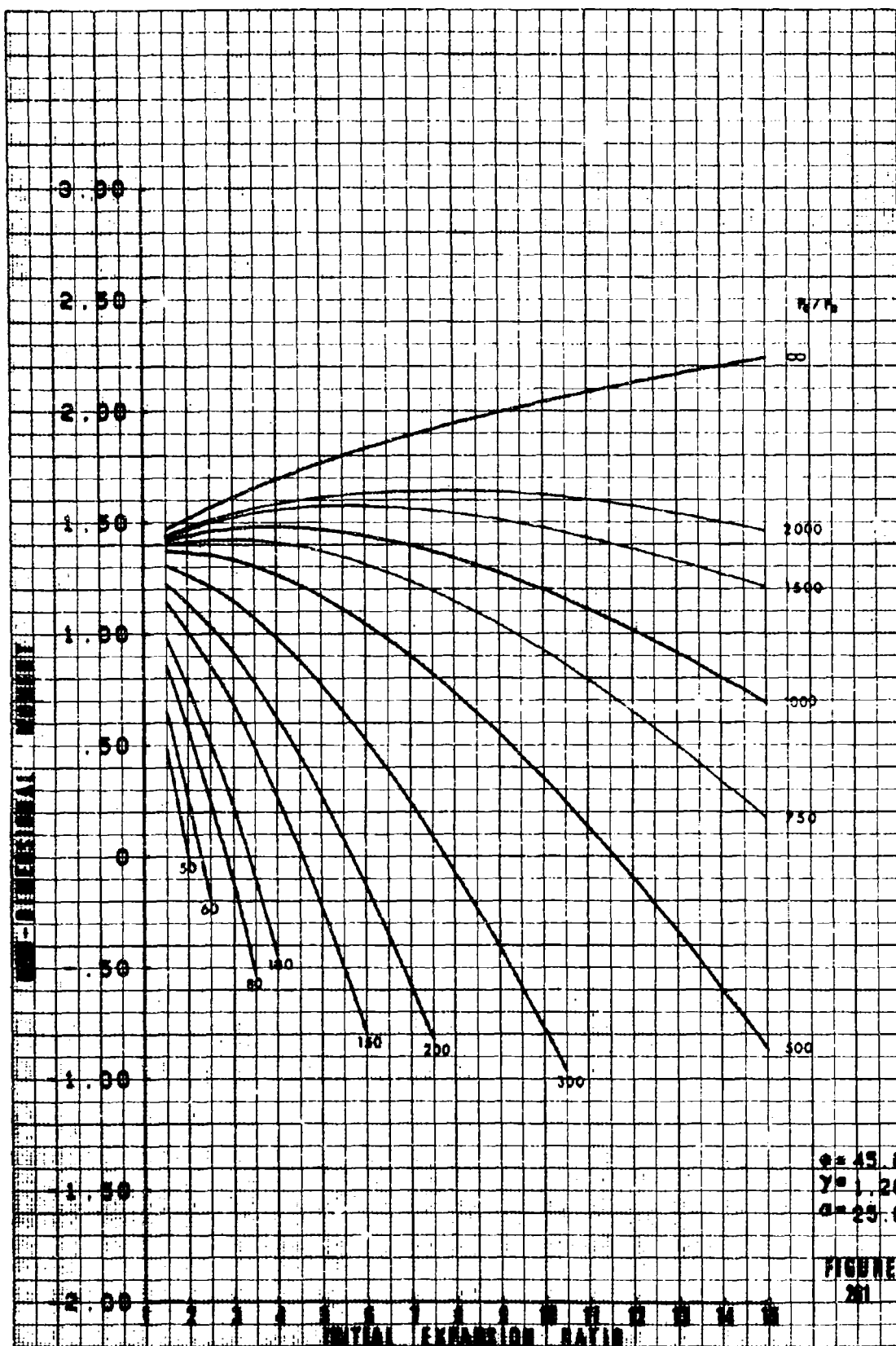
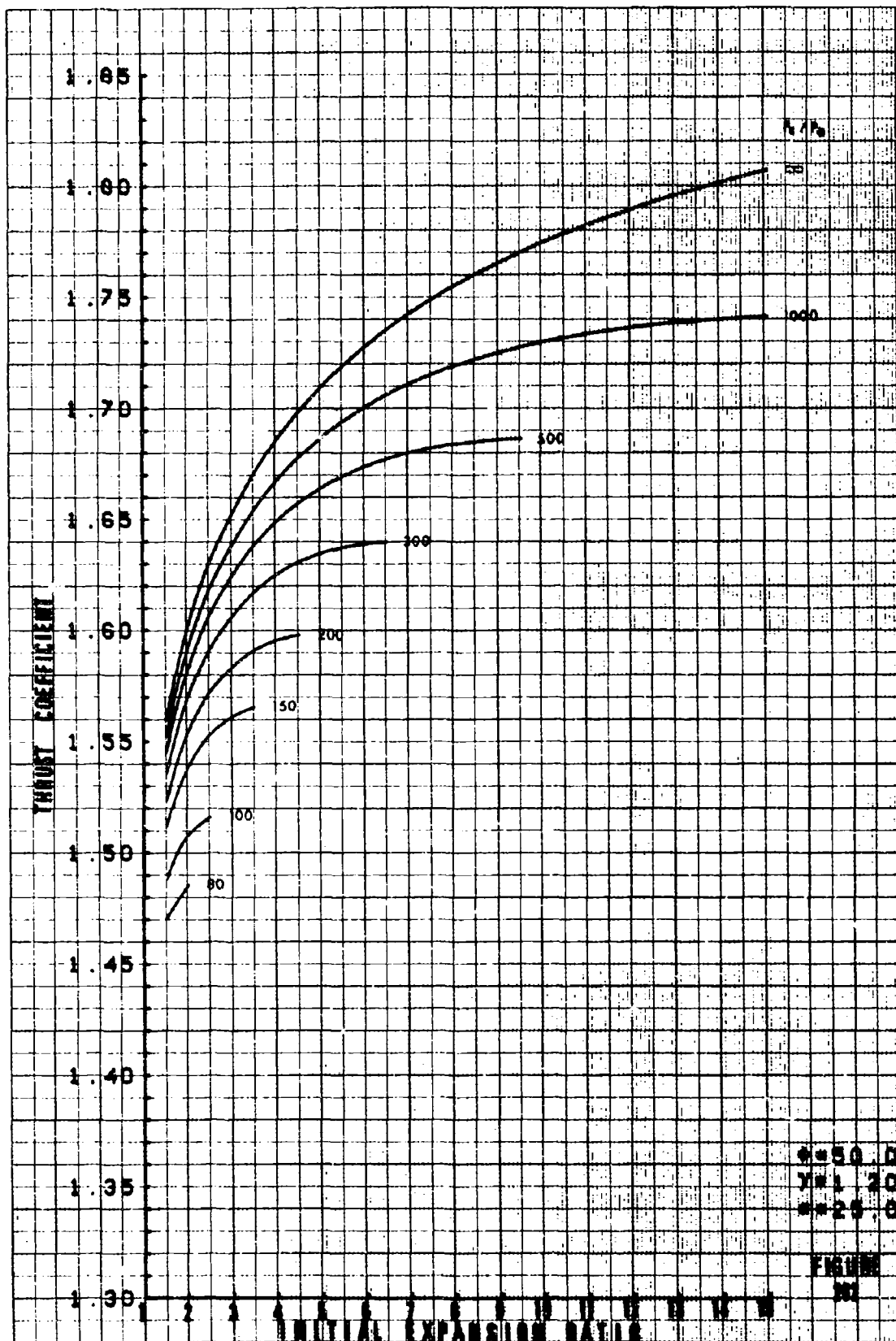
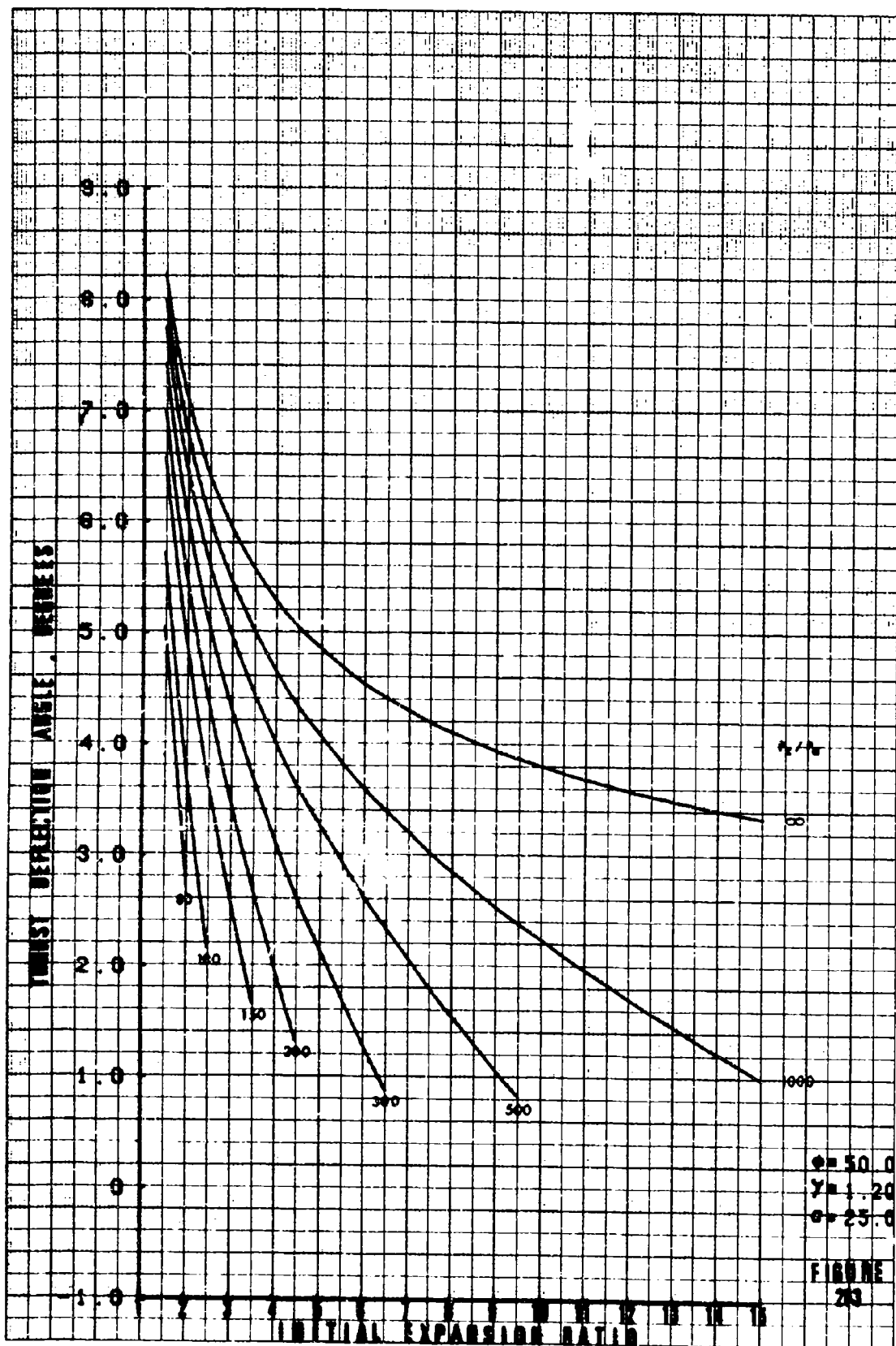
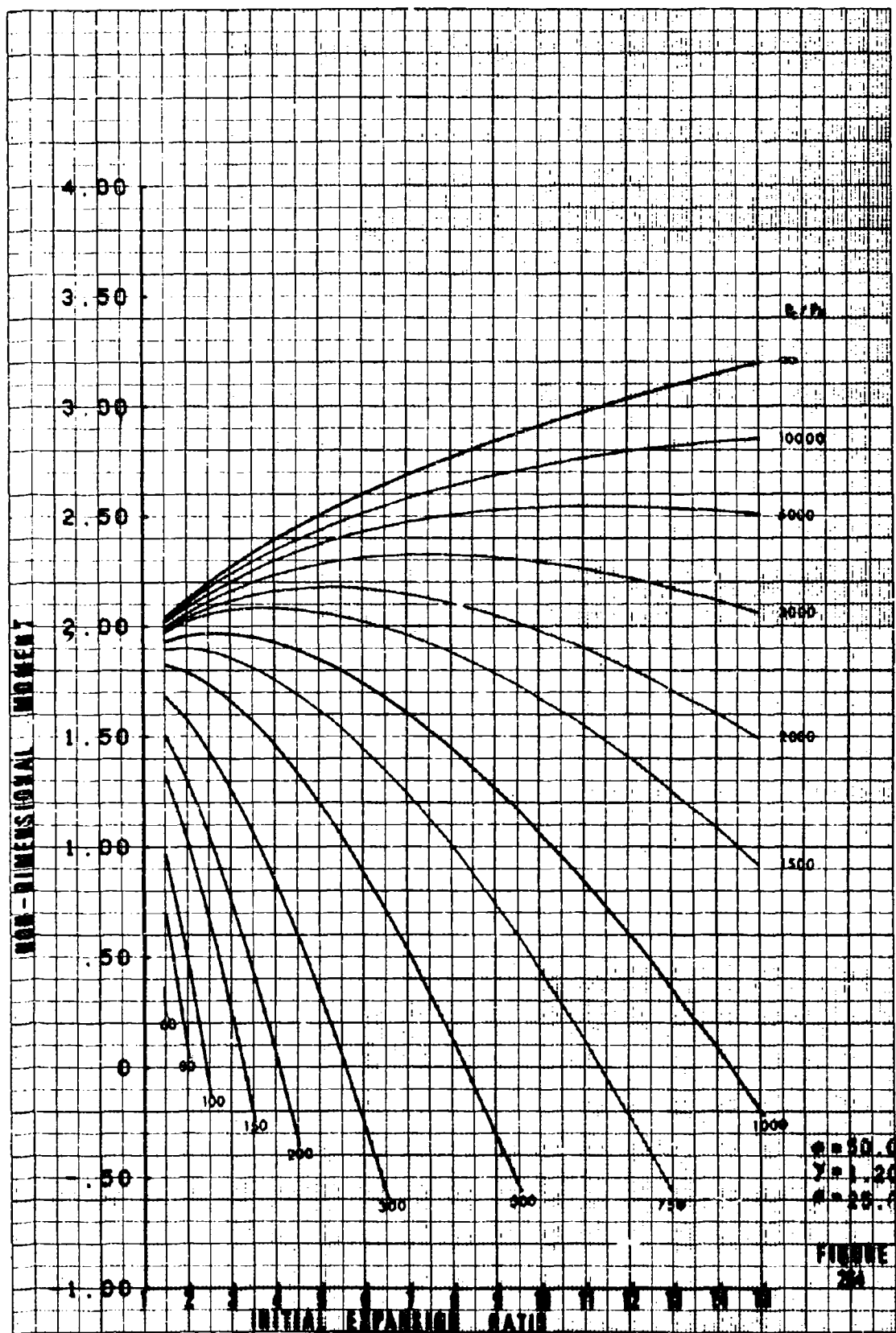


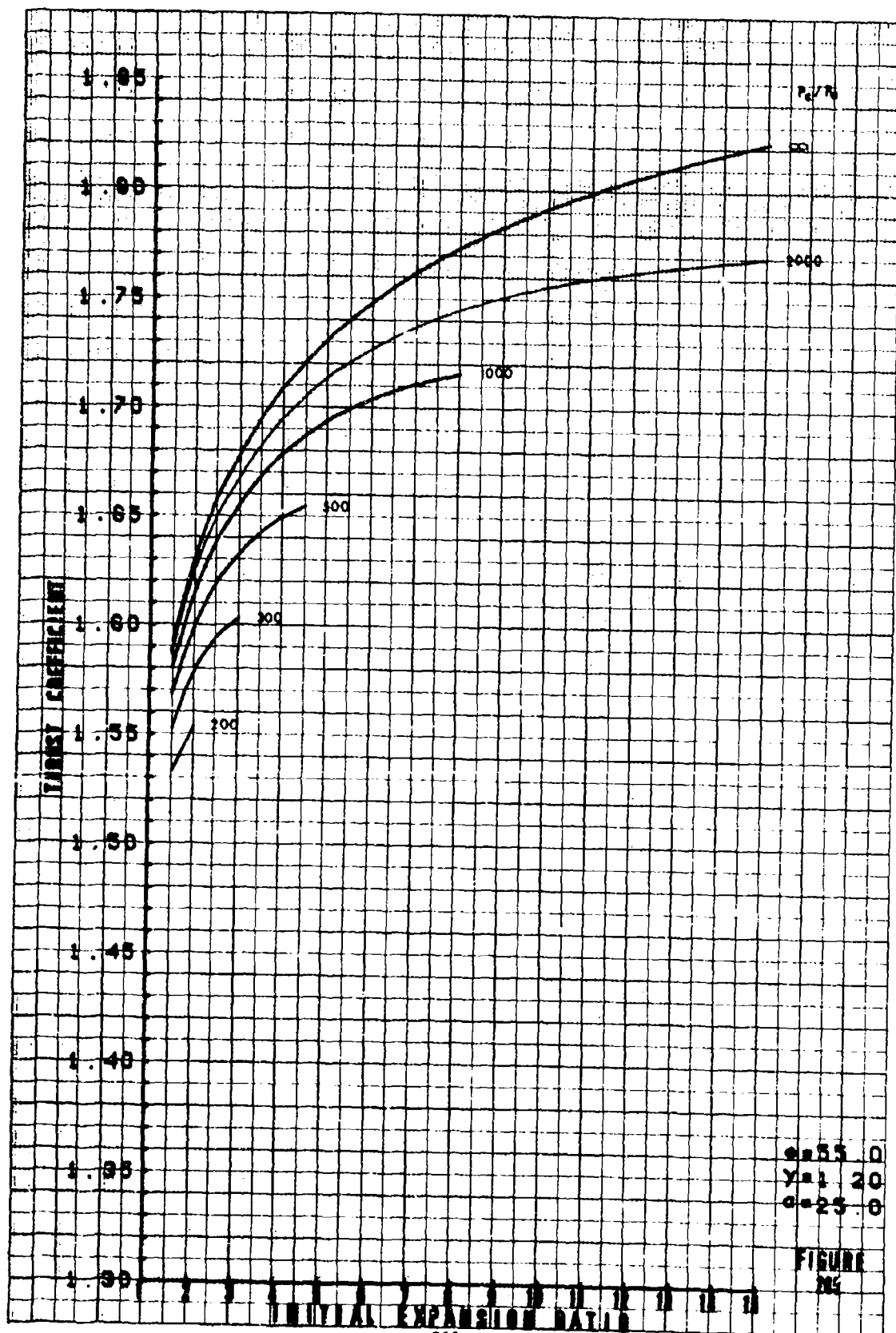
FIGURE 20

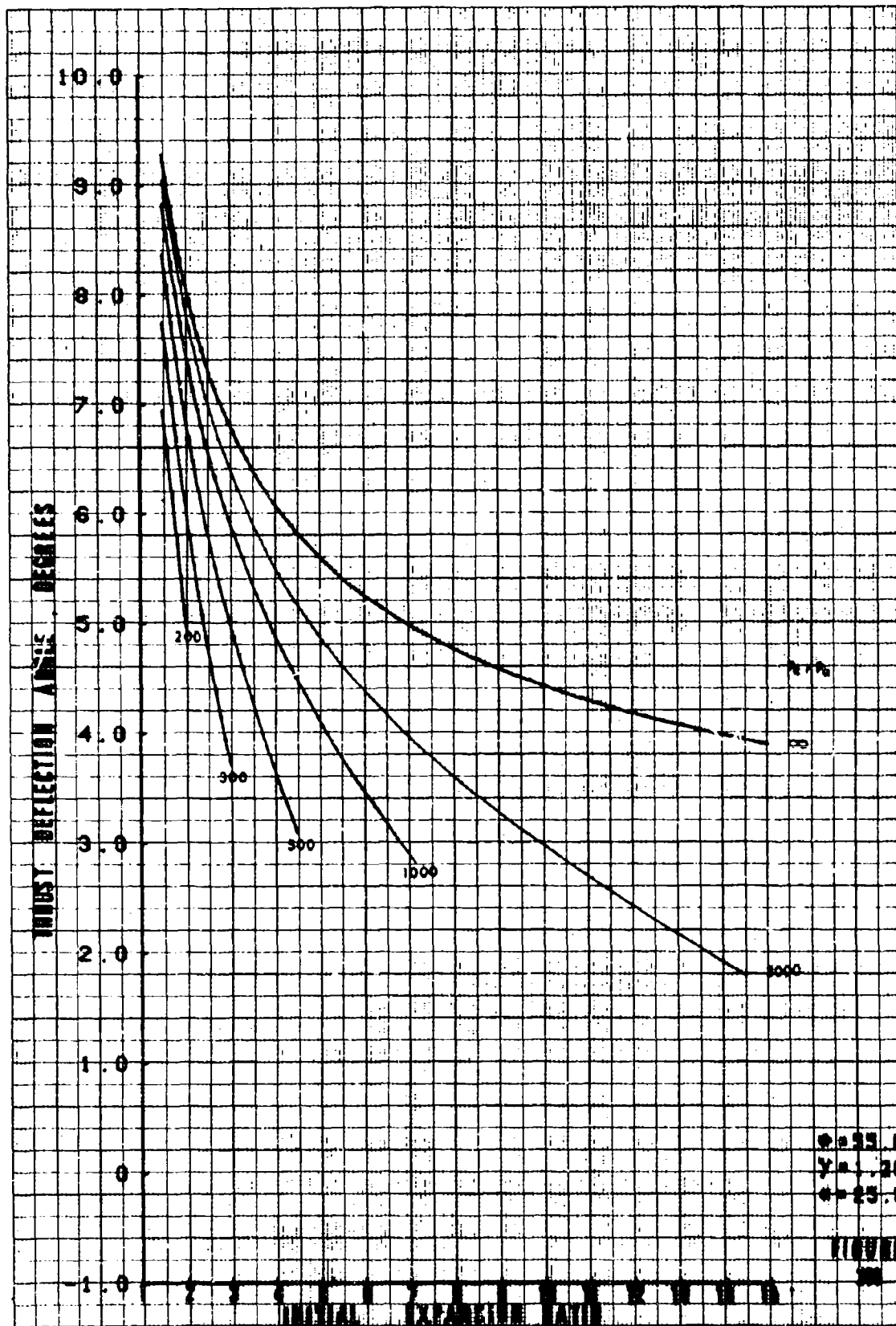


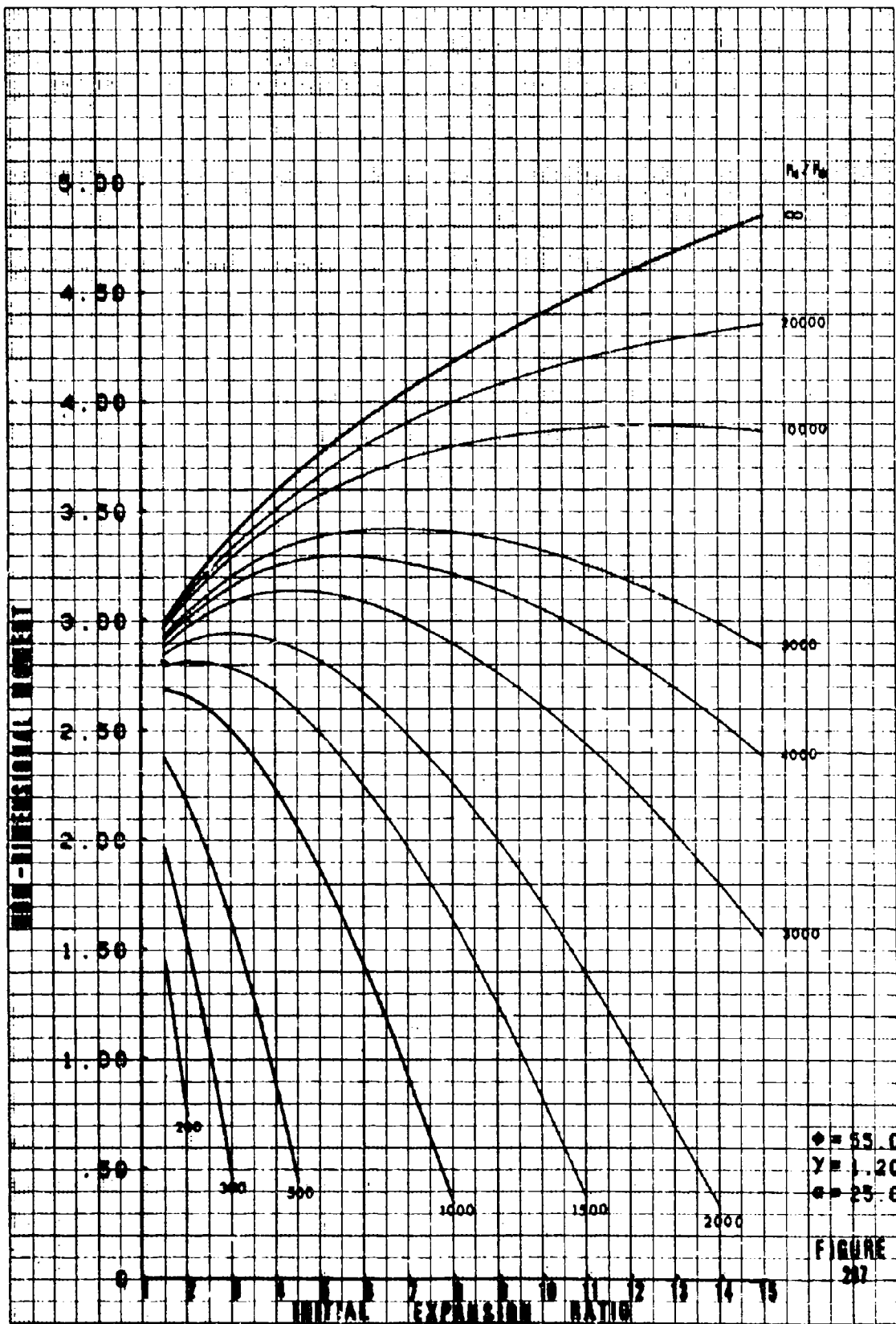


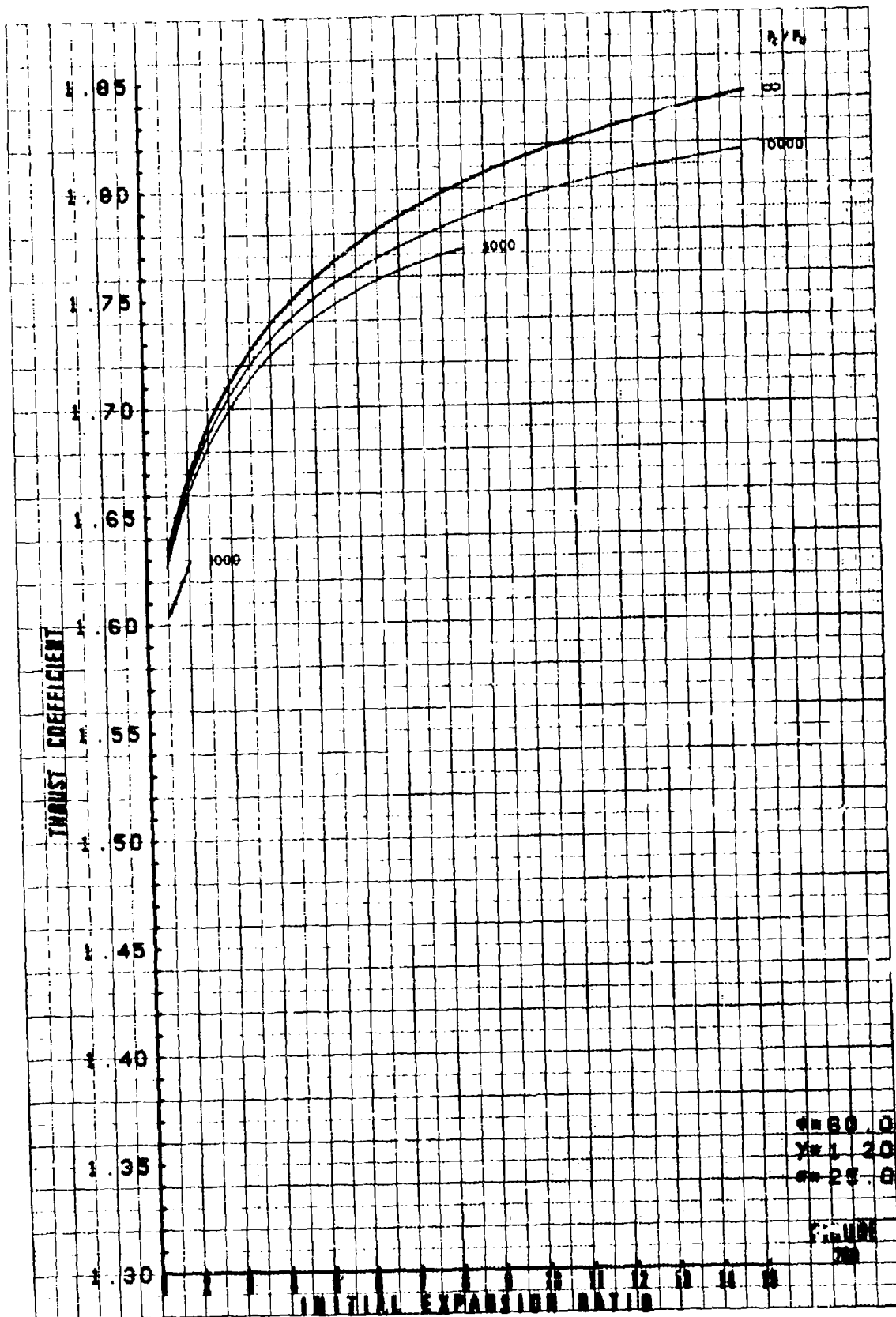












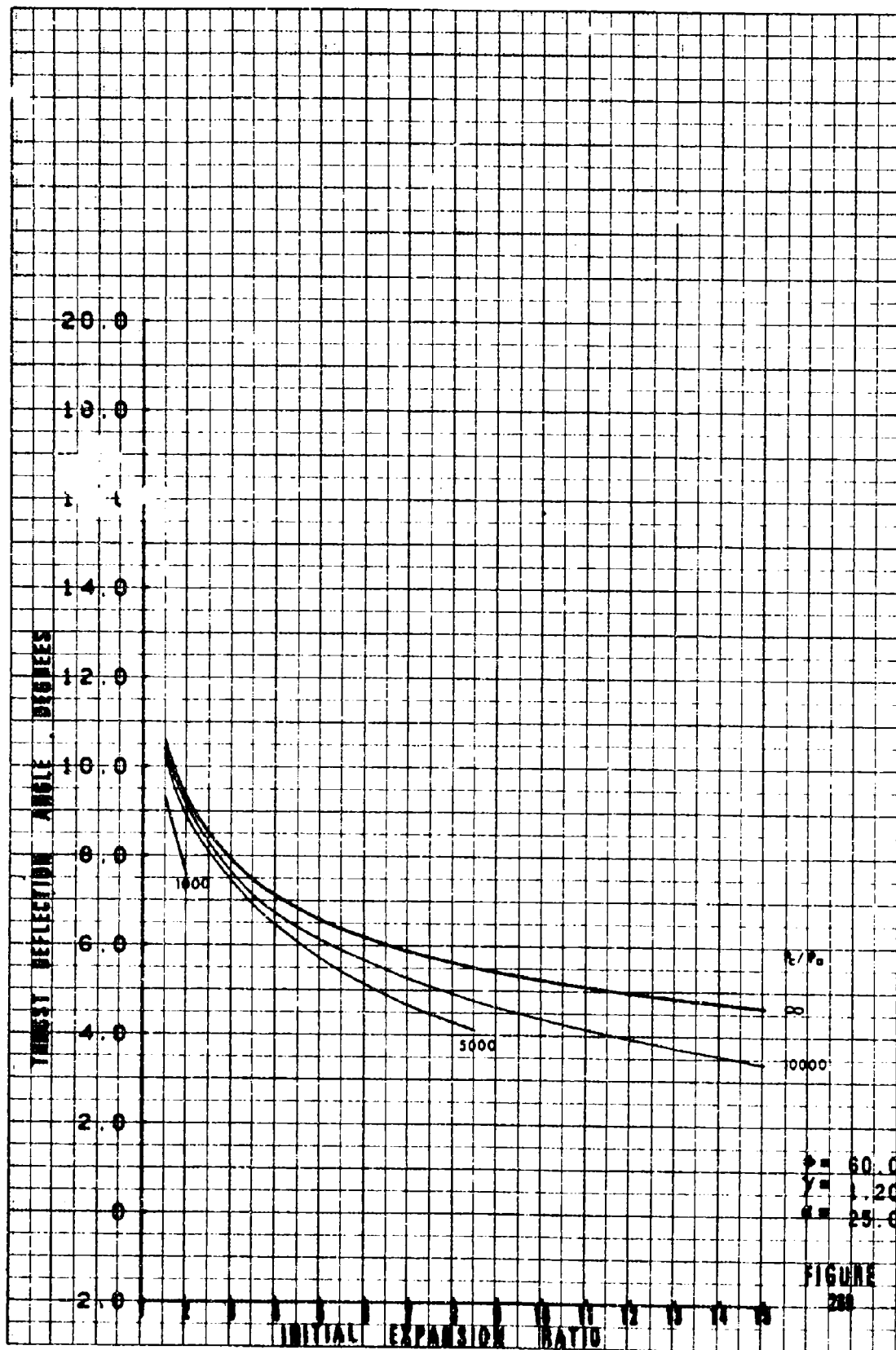
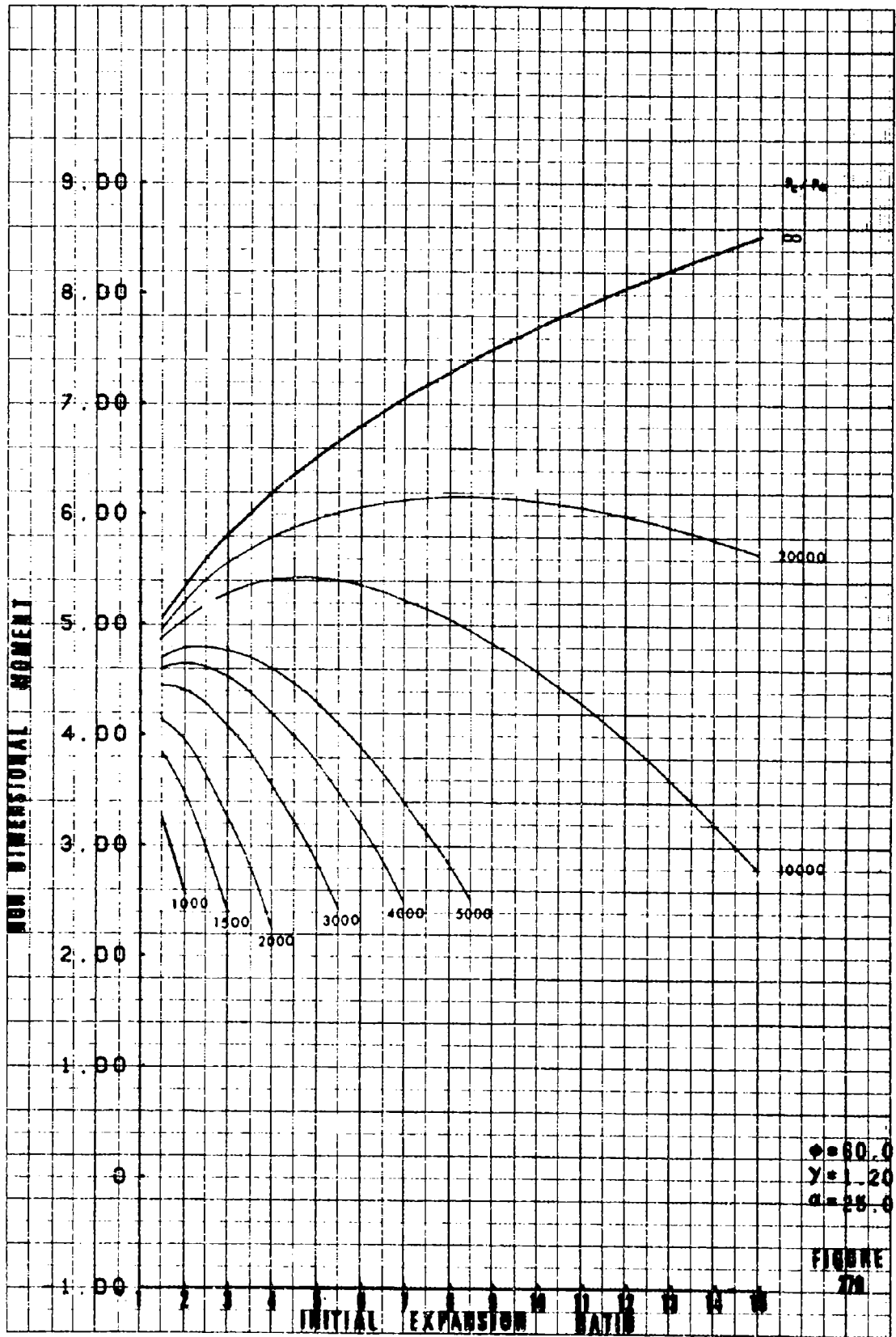
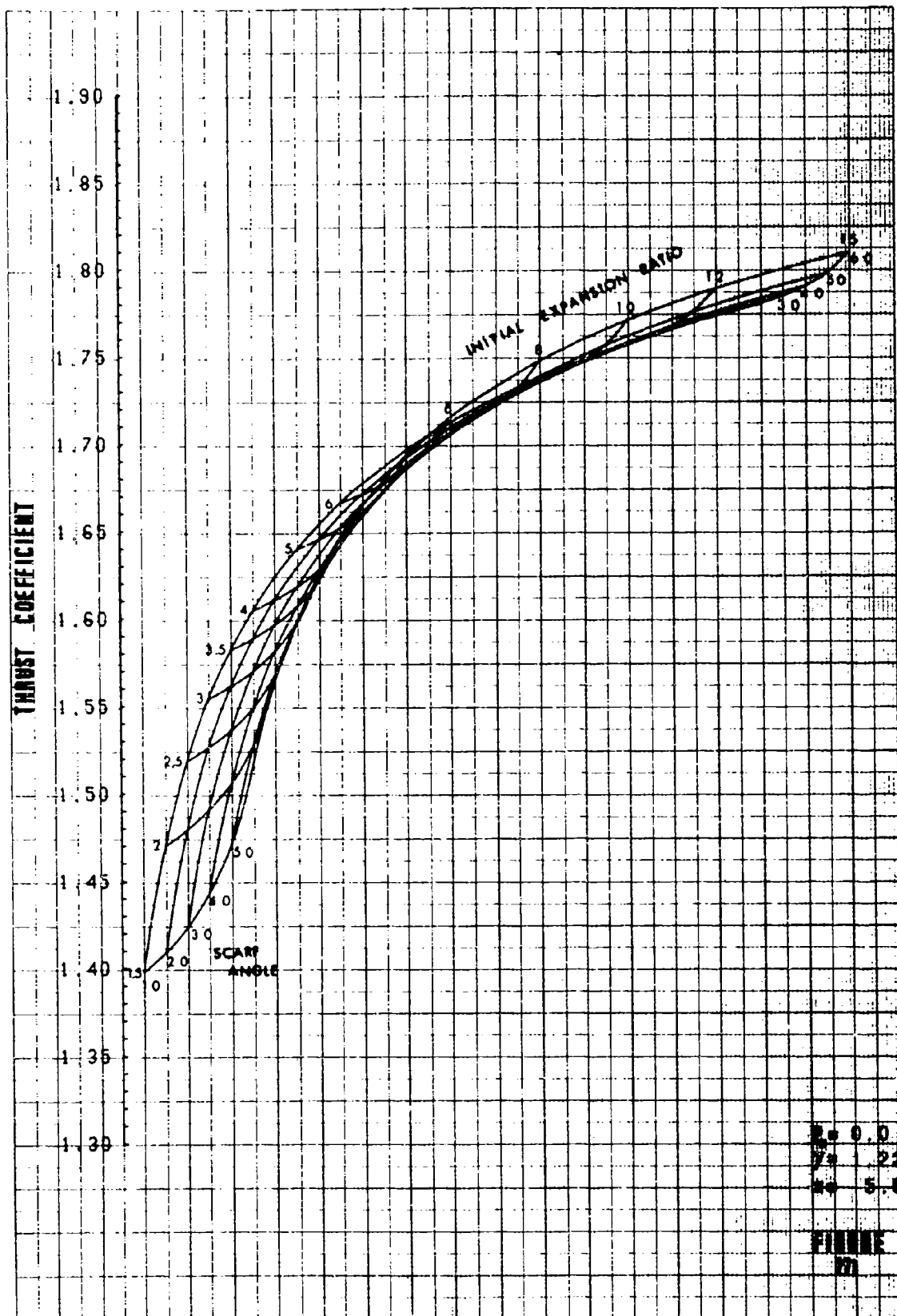
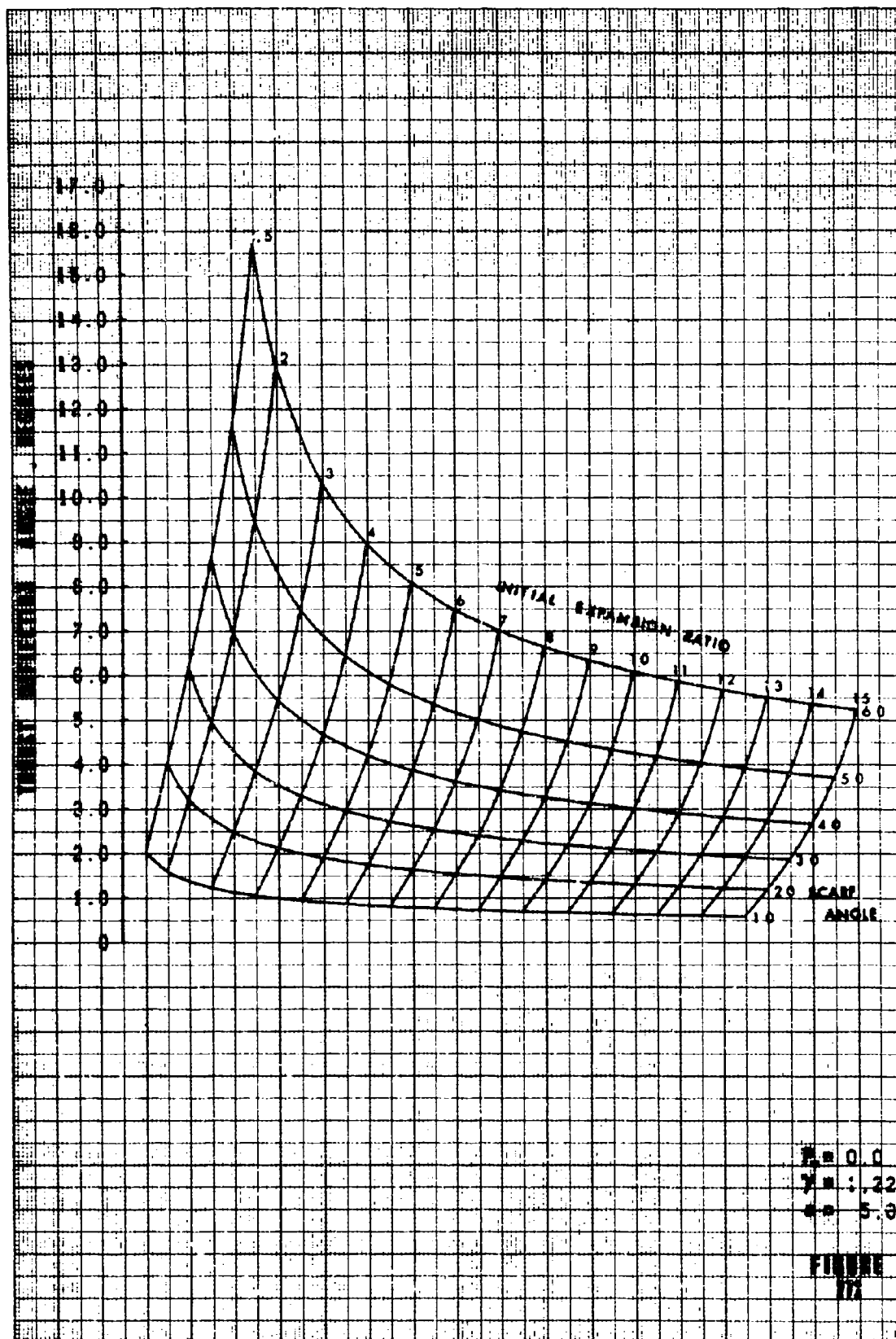


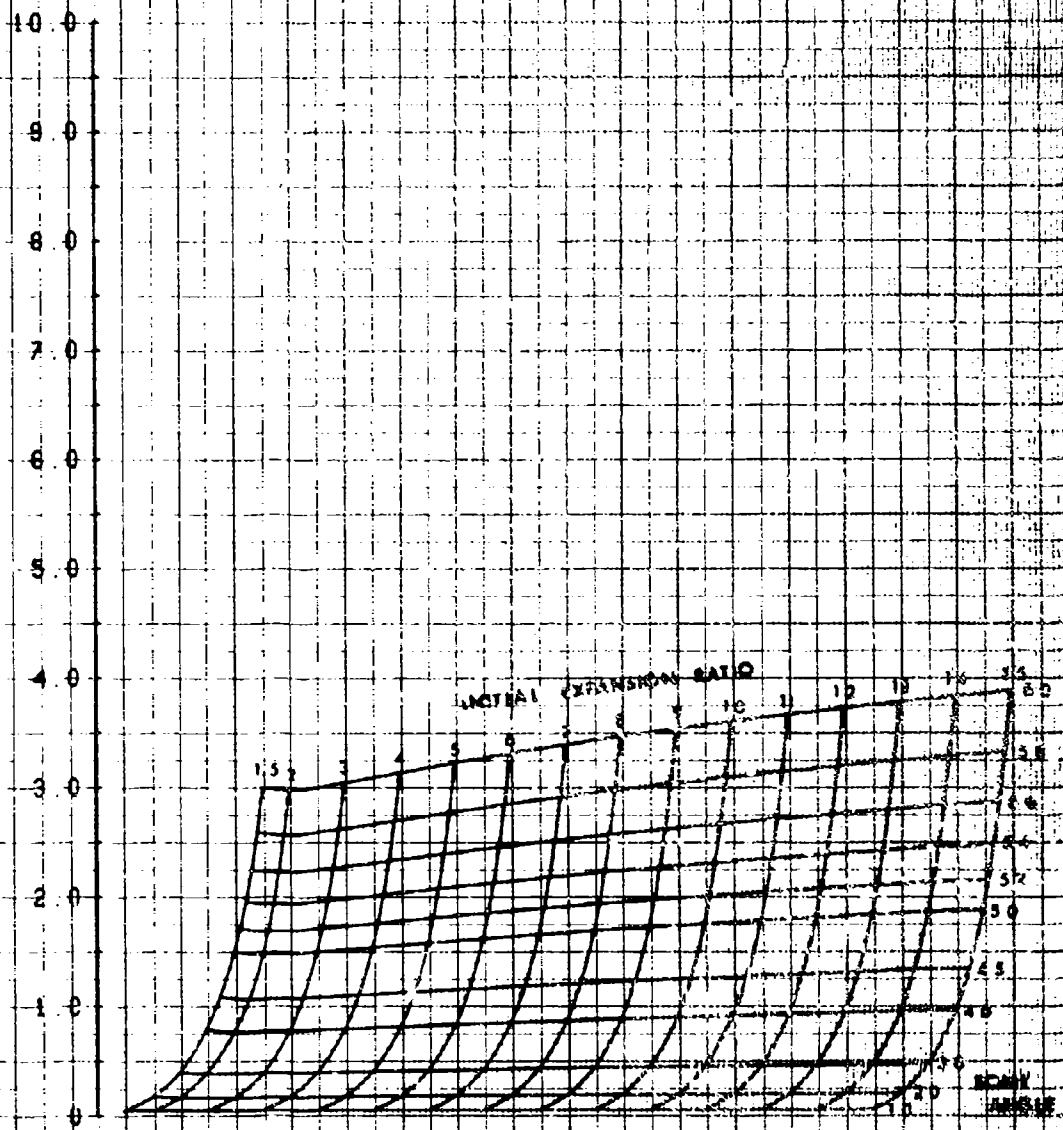
FIGURE 288





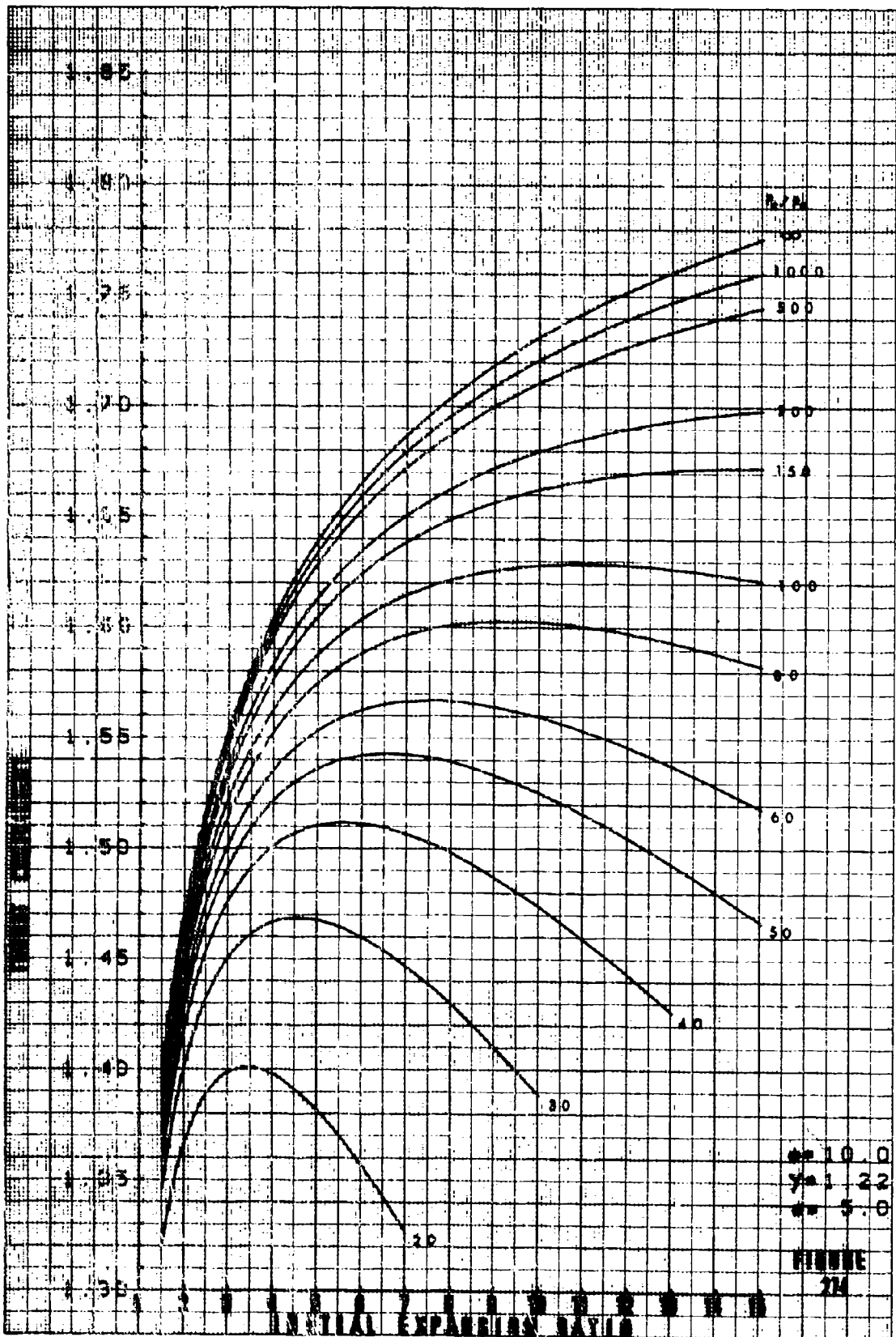


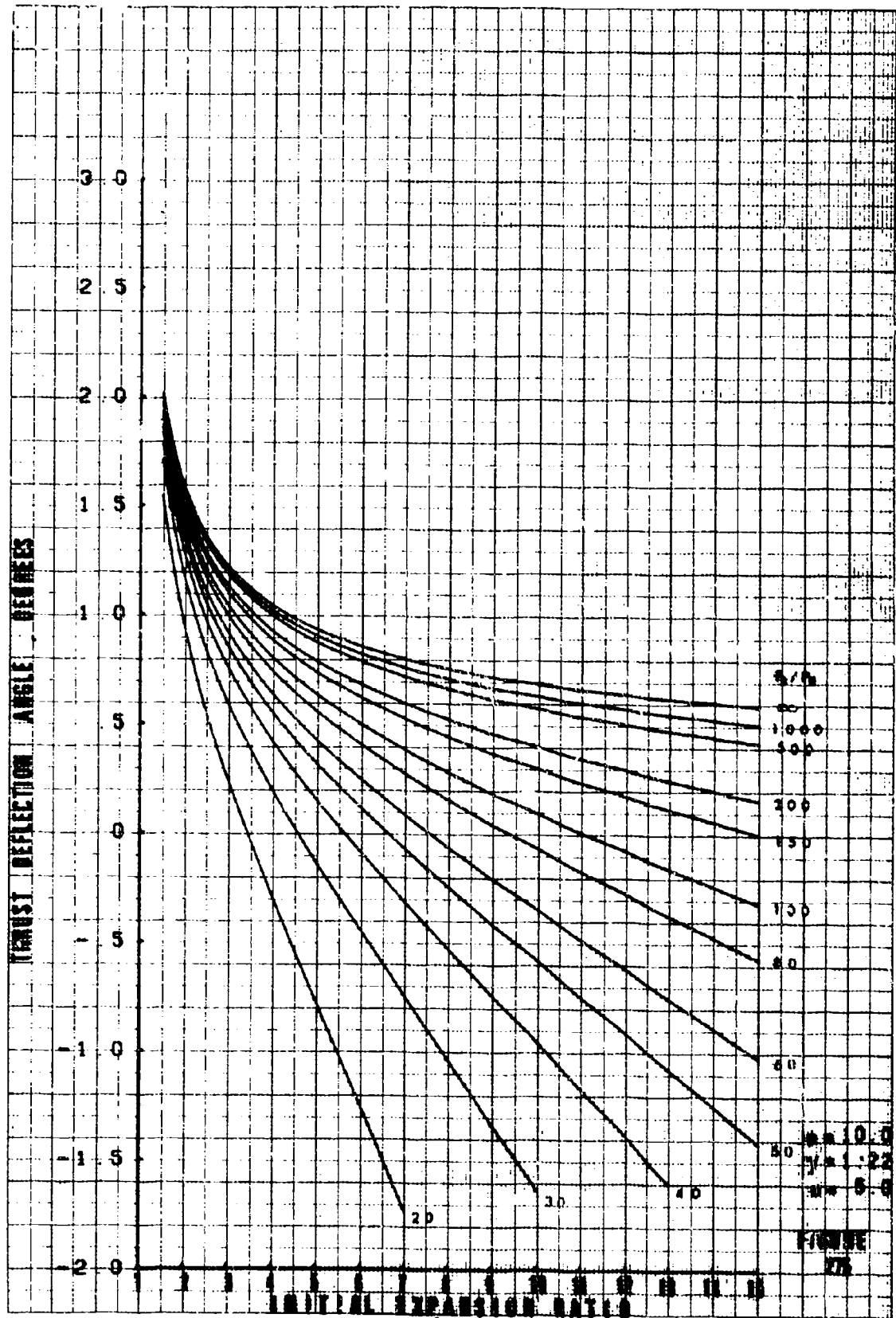
3D-DIMENSIONAL MOMENT

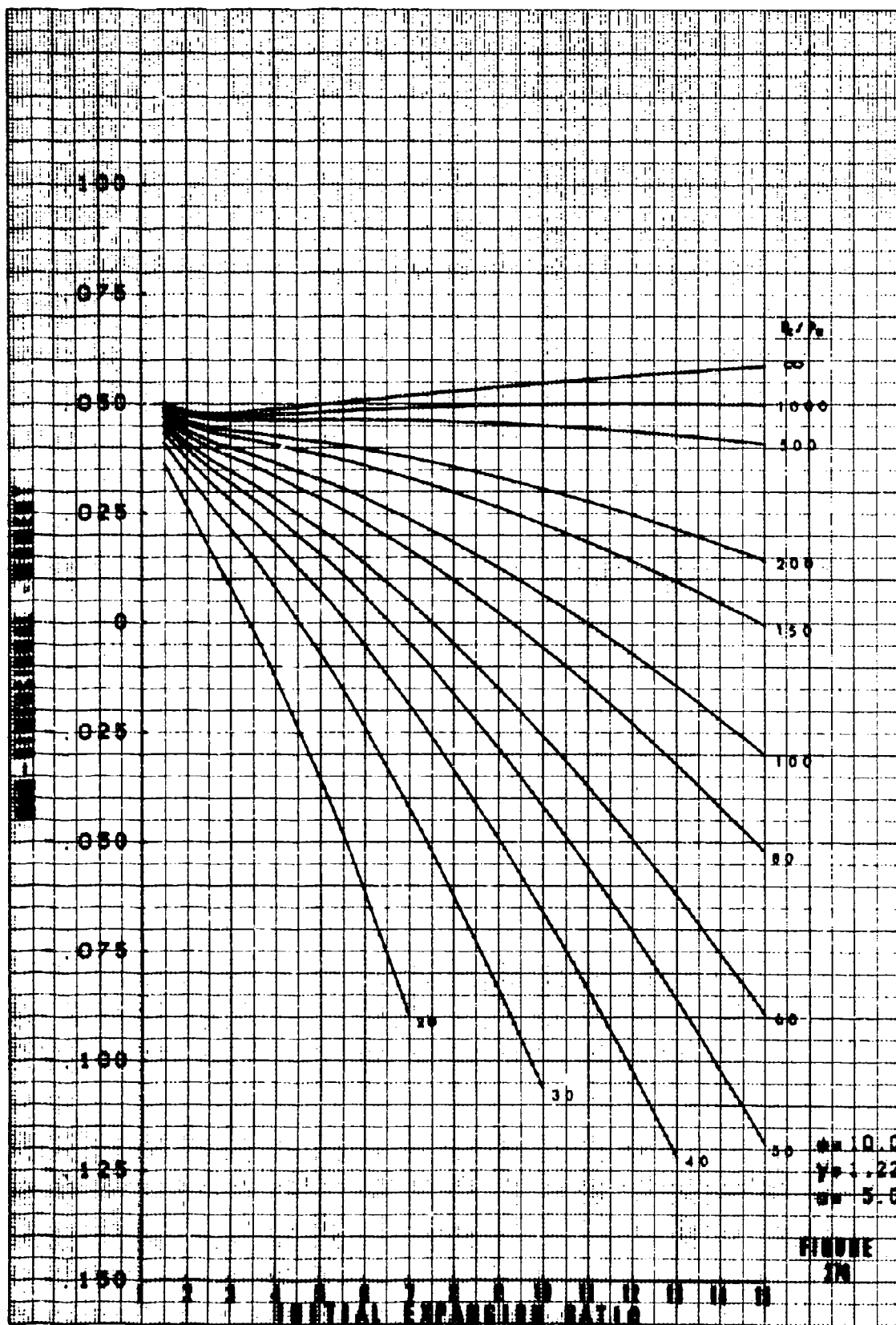


$R = 0.1$
 $T = 0.2$
 $W = 0.3$

FIGURE 10







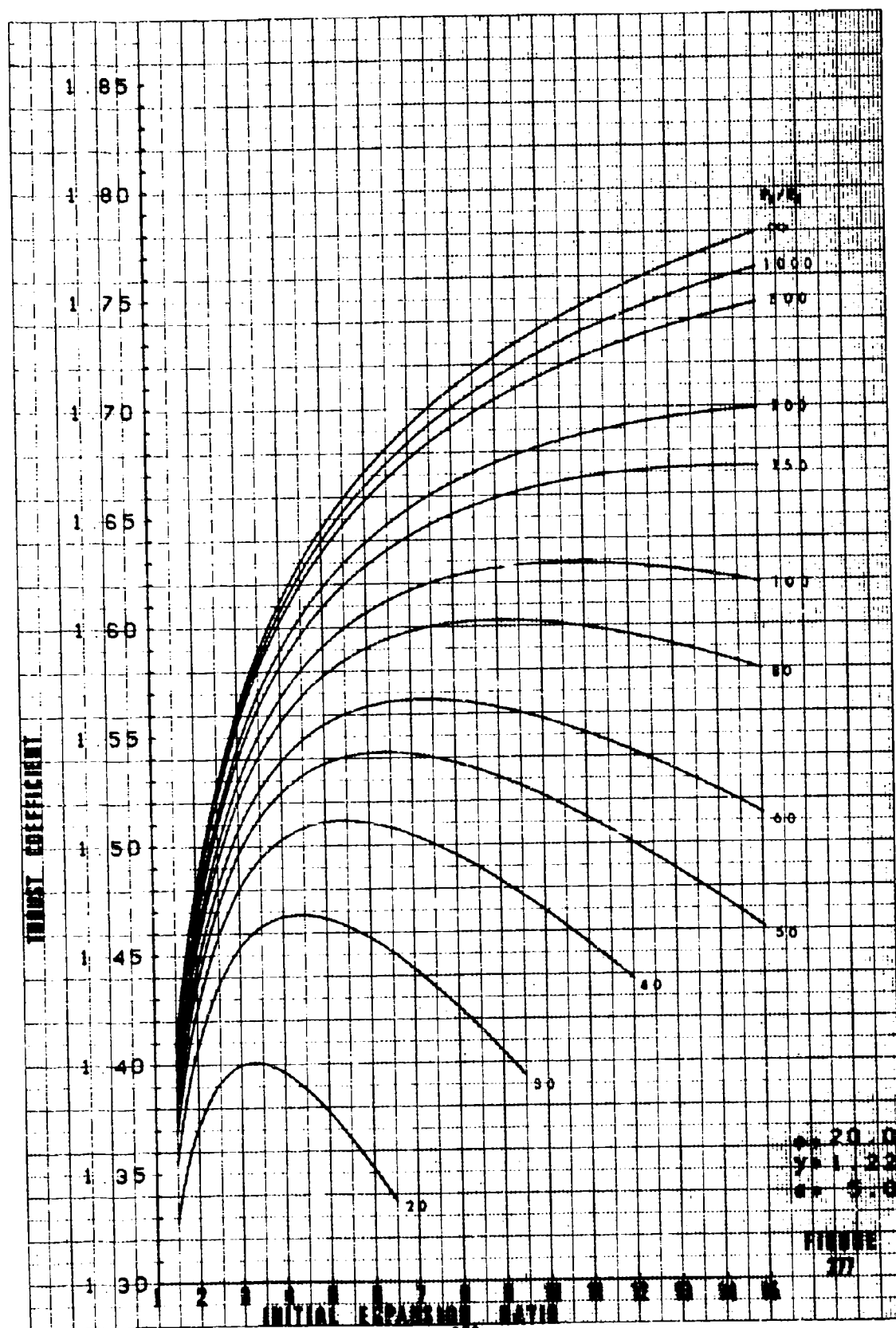
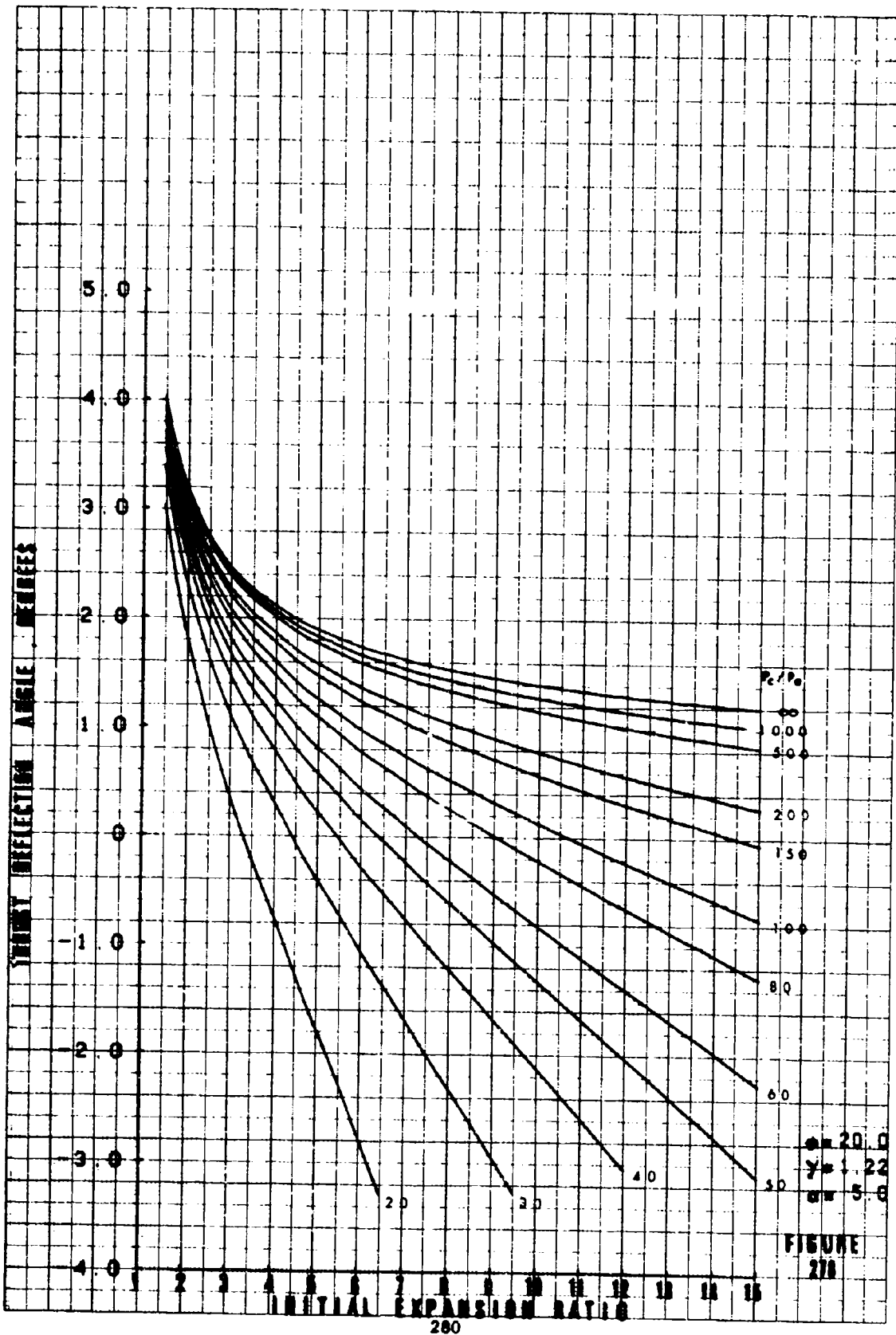
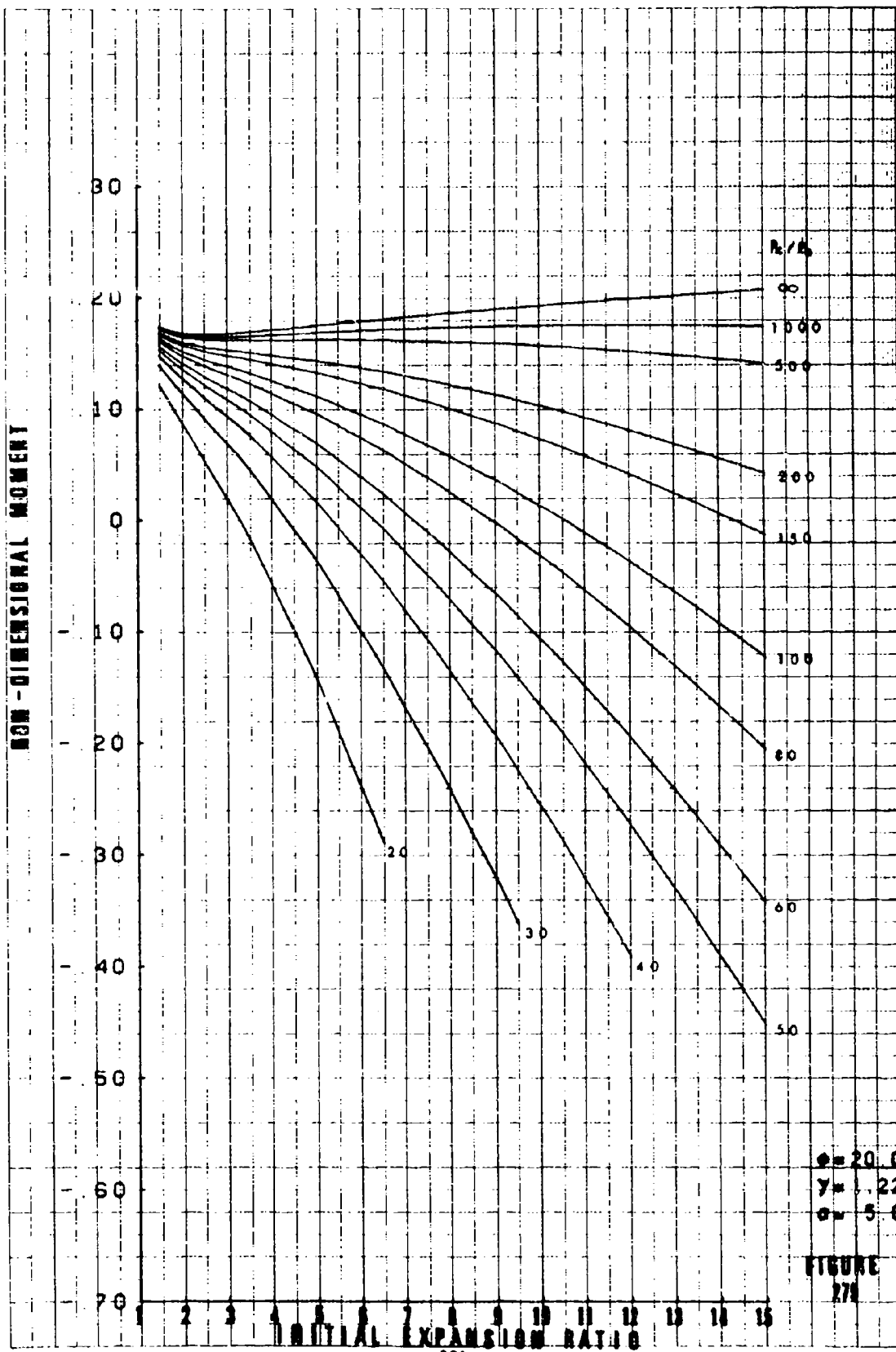
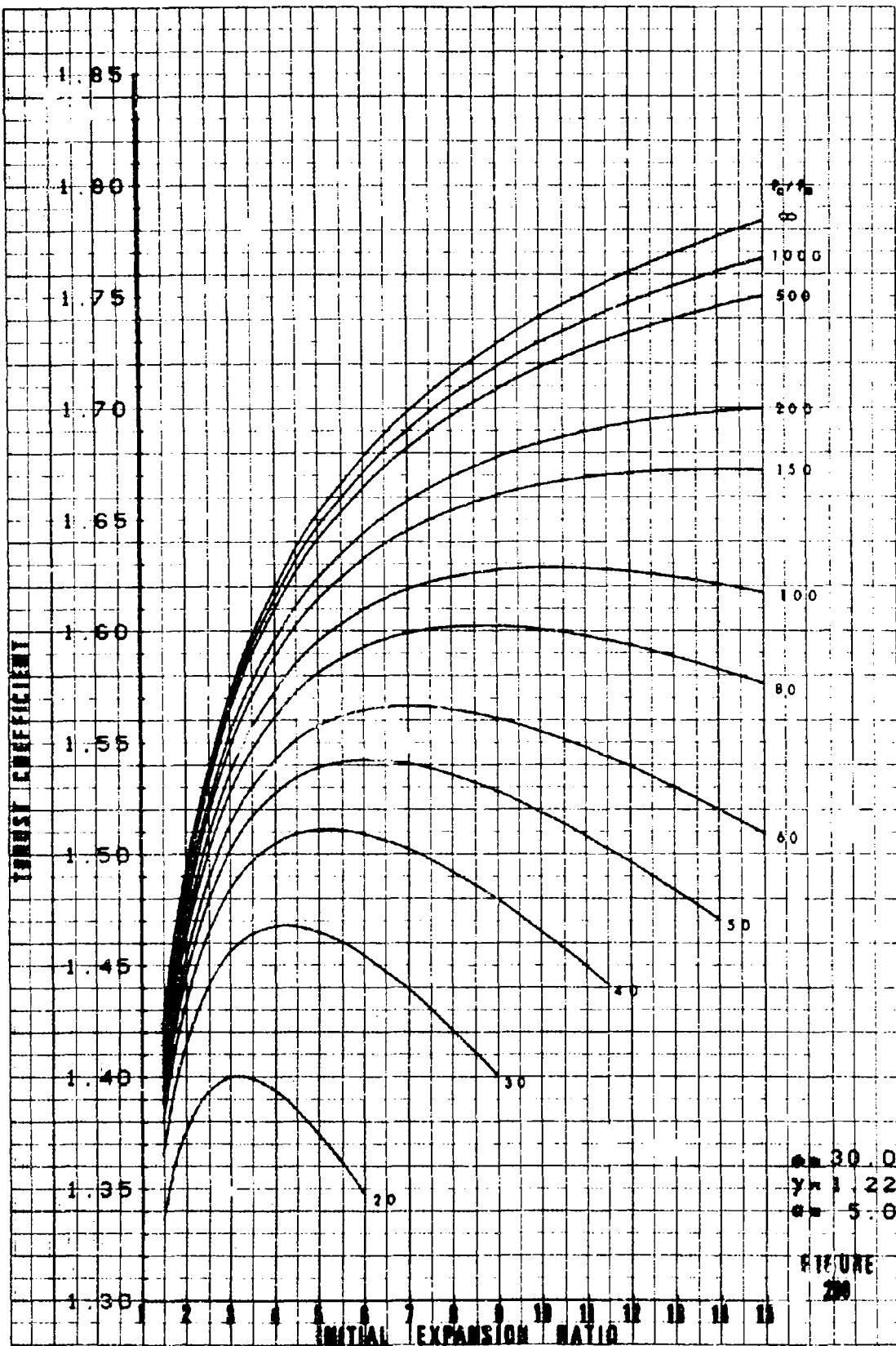


FIGURE 277







$\gamma = 30.0$
 $\gamma = 1.22$
 $\gamma = 5.0$

FIGURE 200

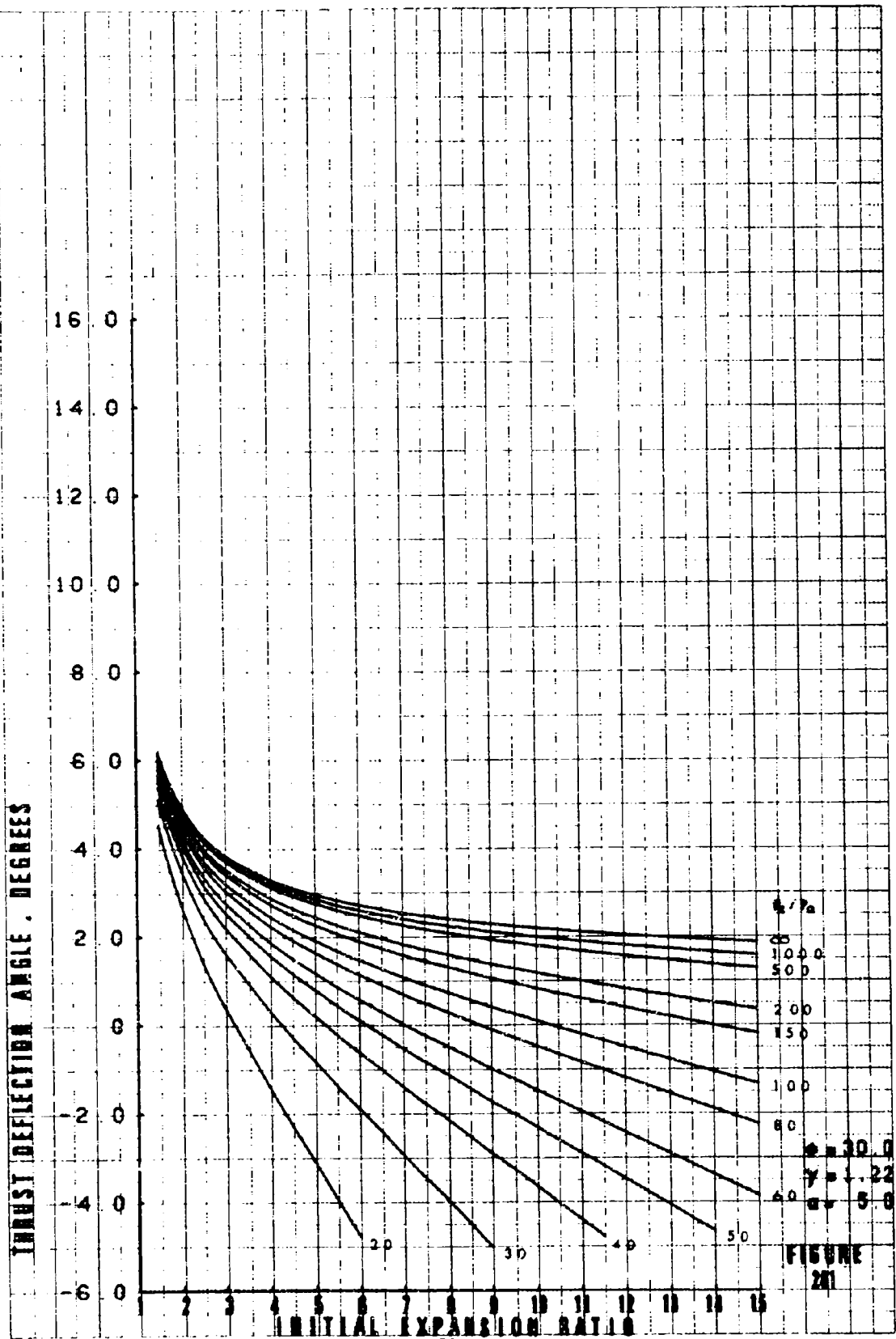
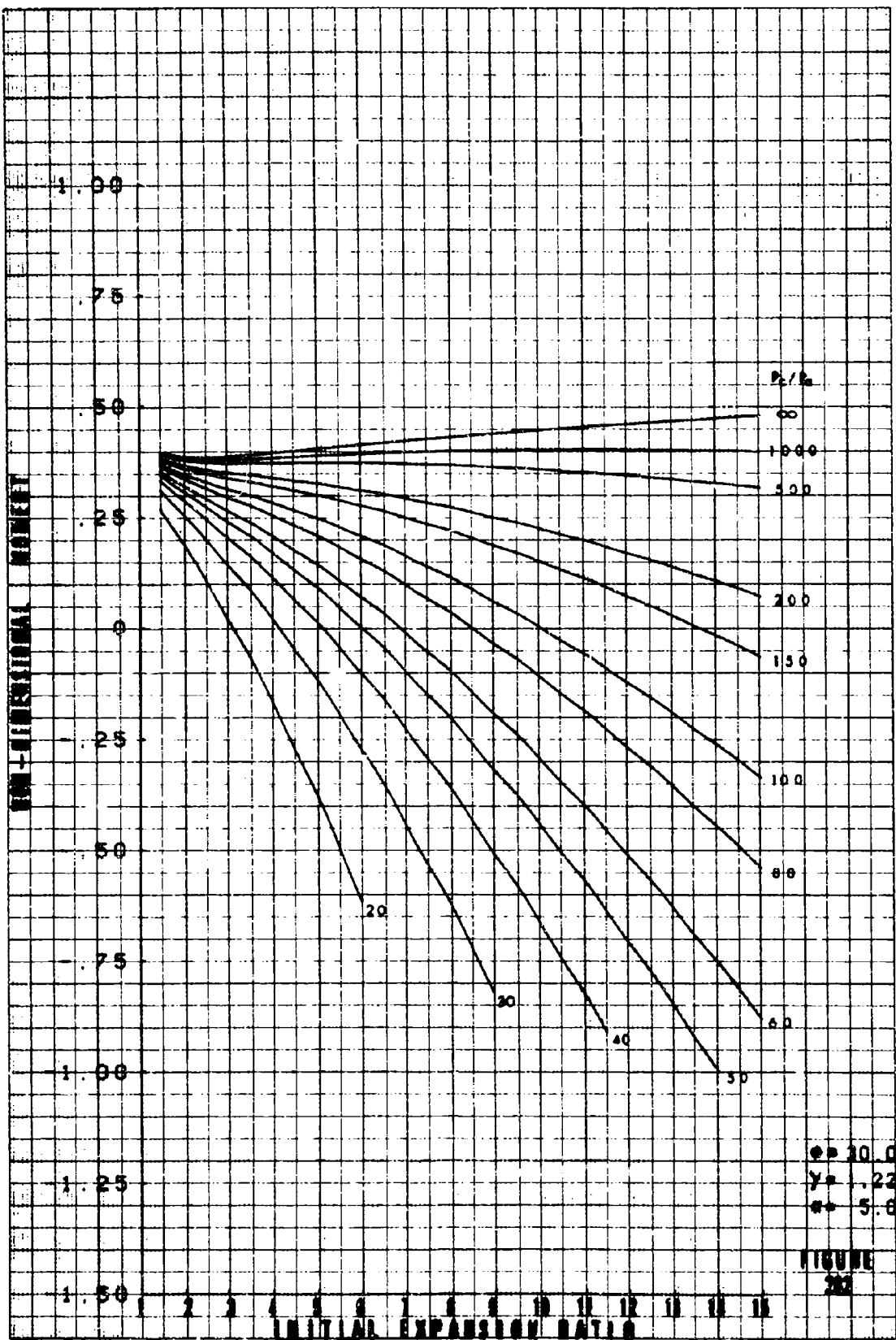
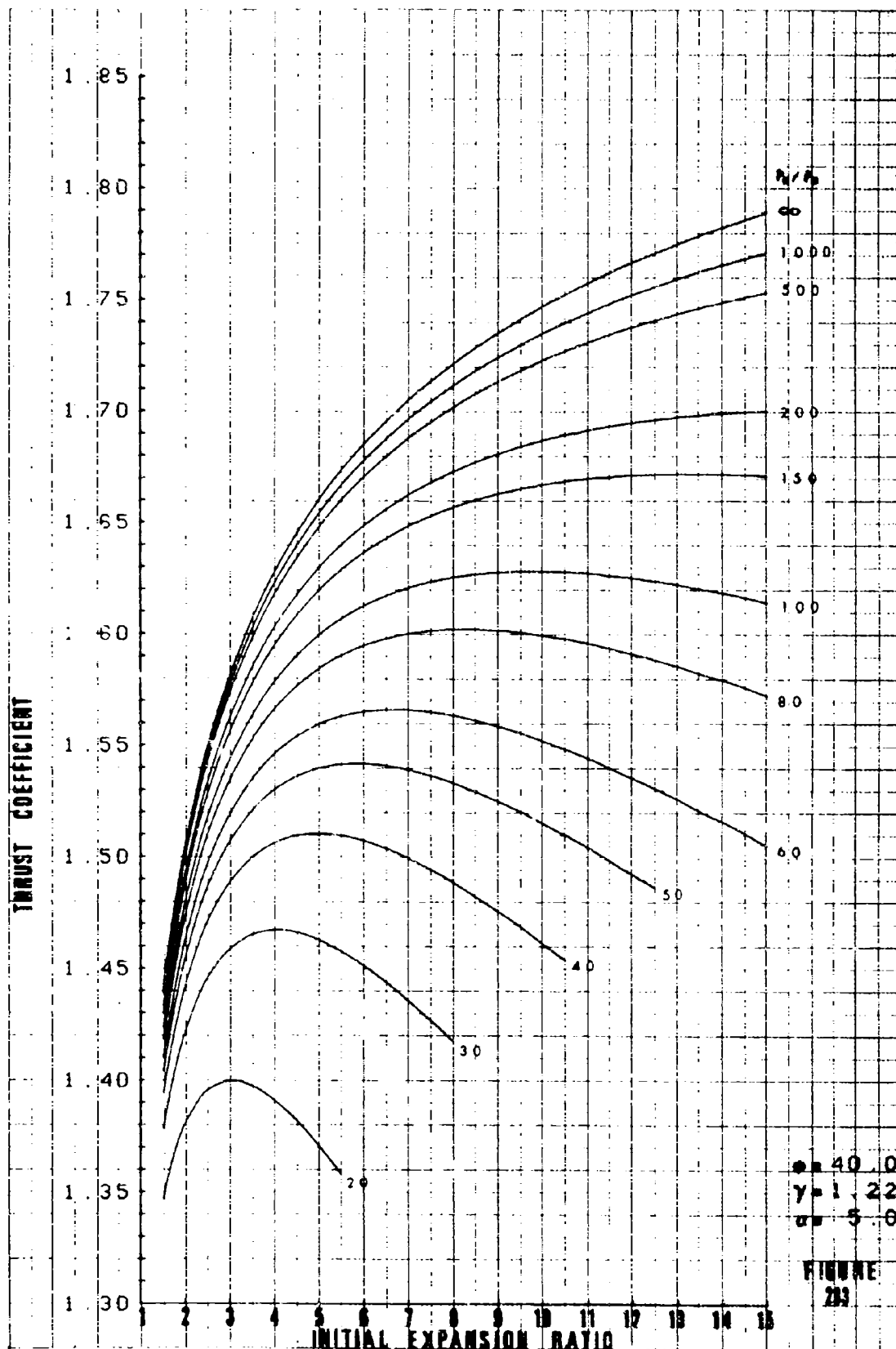


FIGURE 201





$\gamma = 1.22$
 $\sigma = 5.0$

FIGURE 283

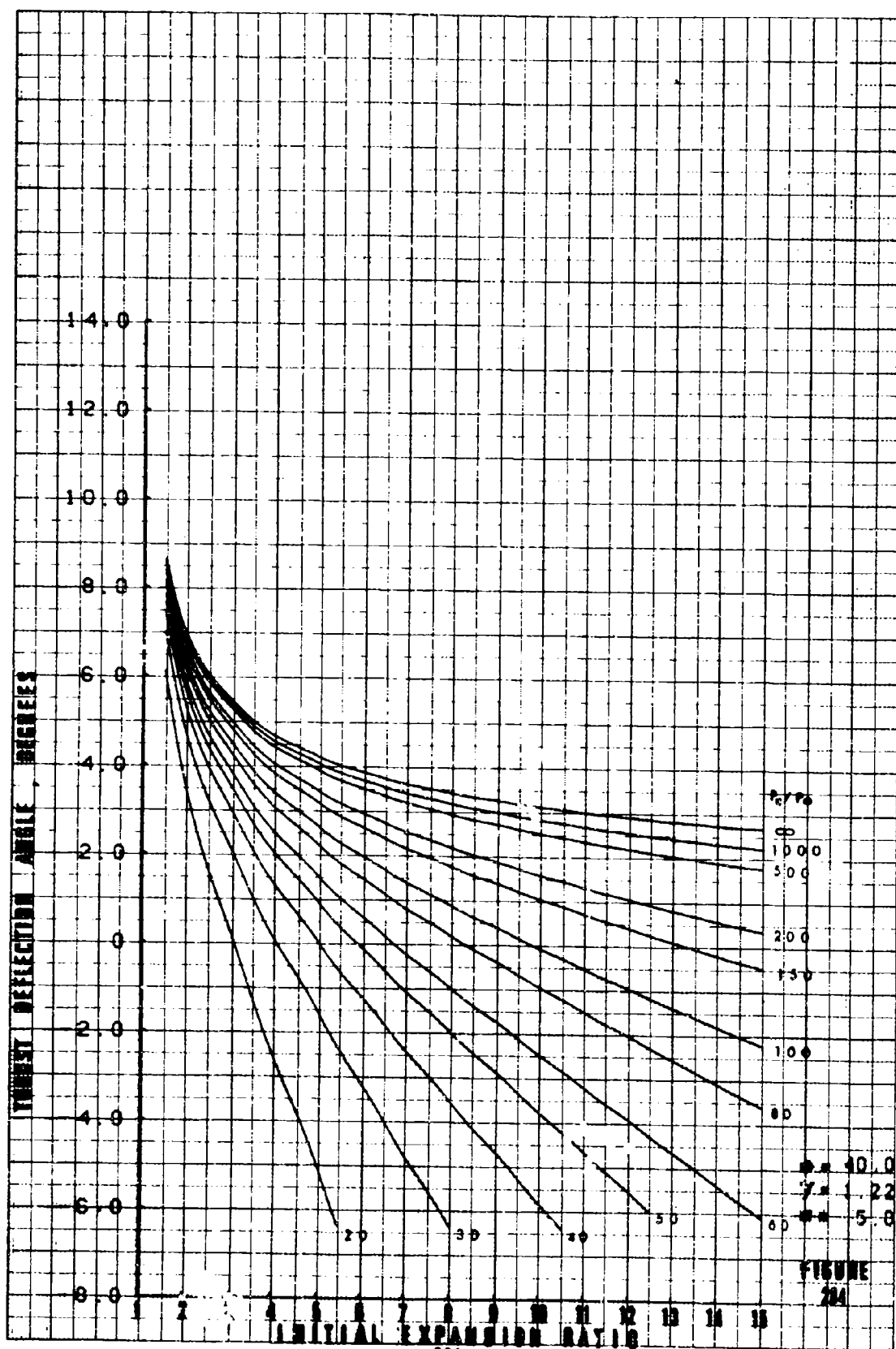
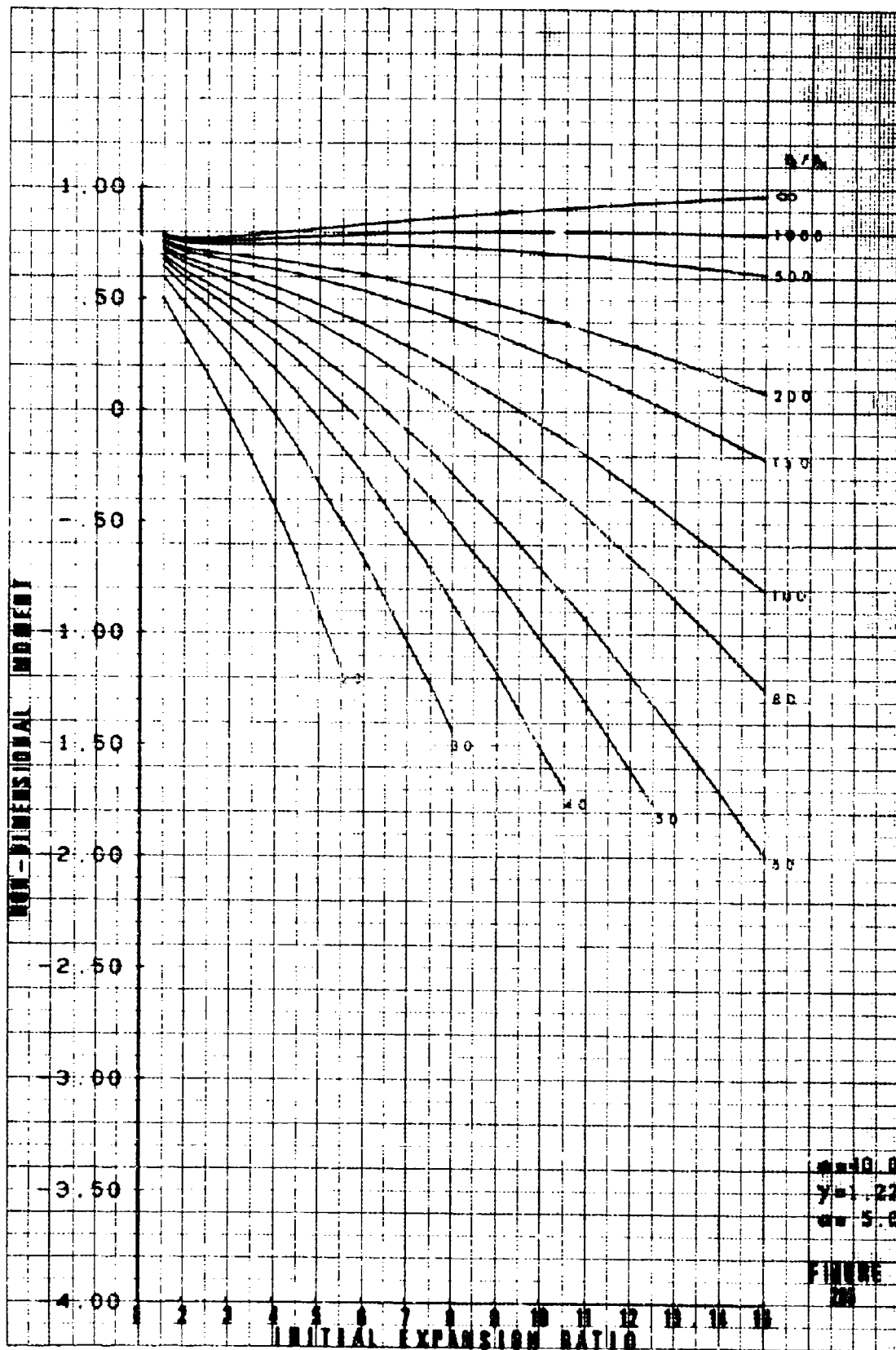
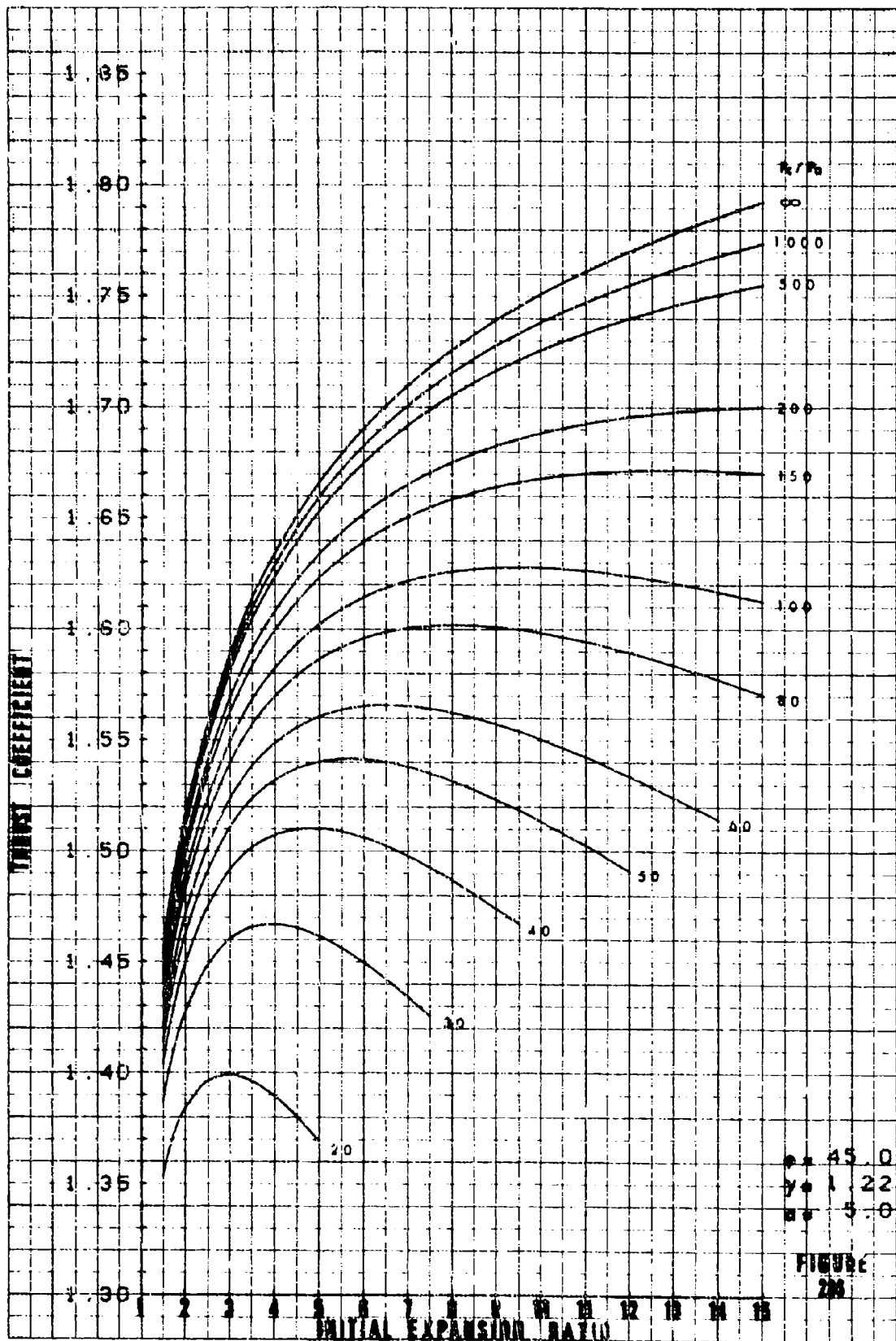


FIGURE 284





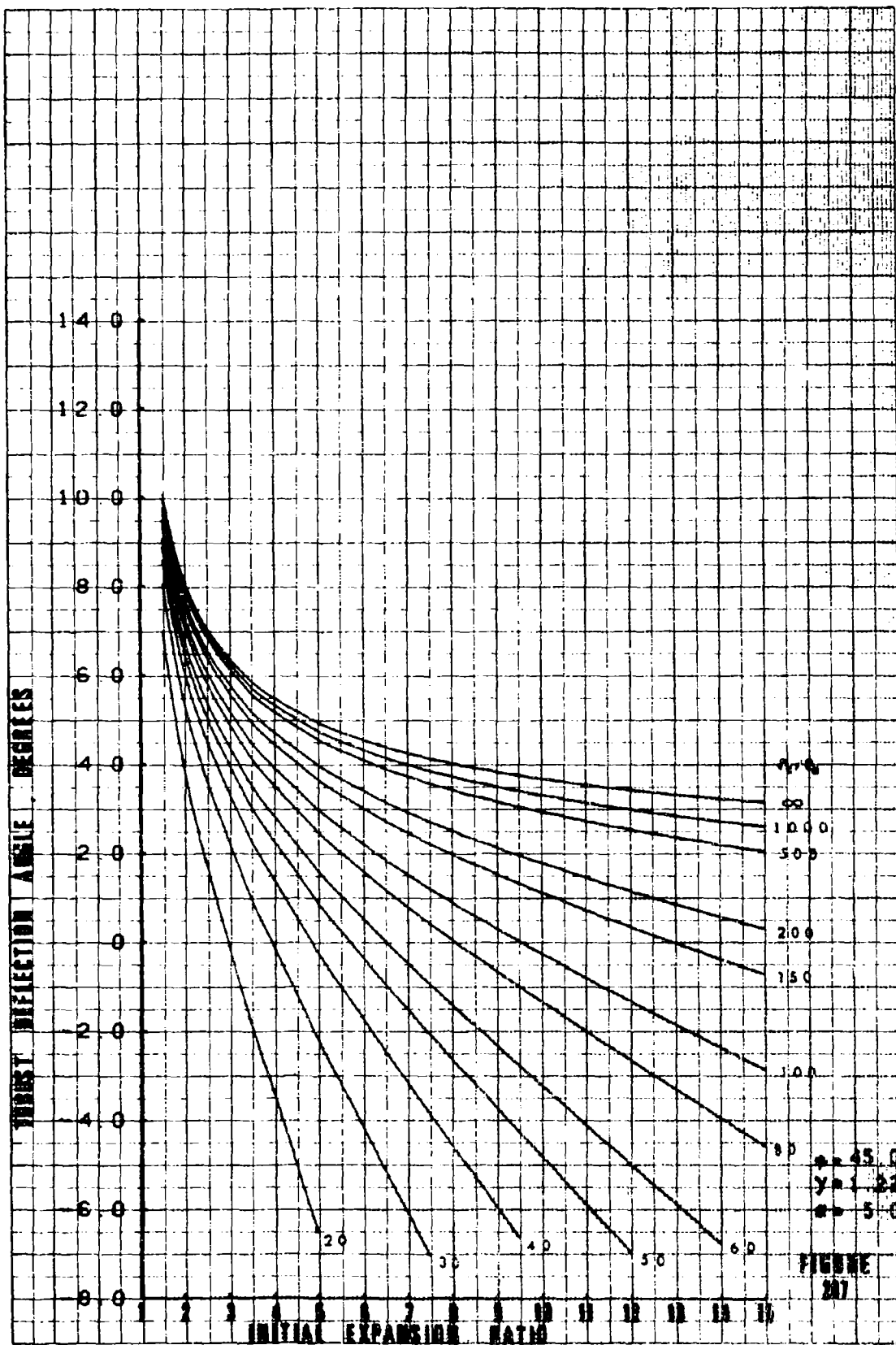
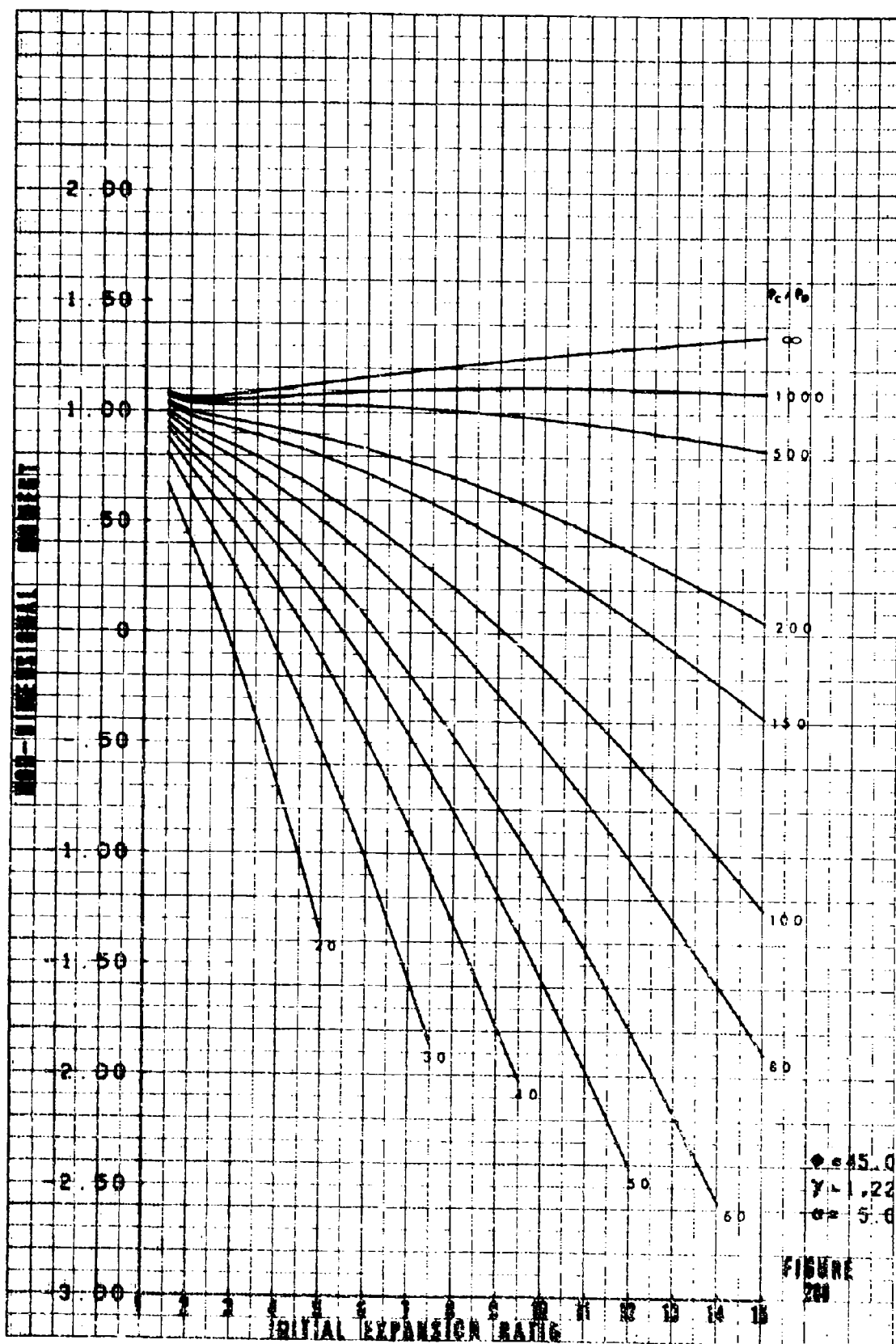


FIGURE 207



$\phi = 45.0$
 $\gamma = 1.22$
 $\sigma = 5.6$

FIGURE 289

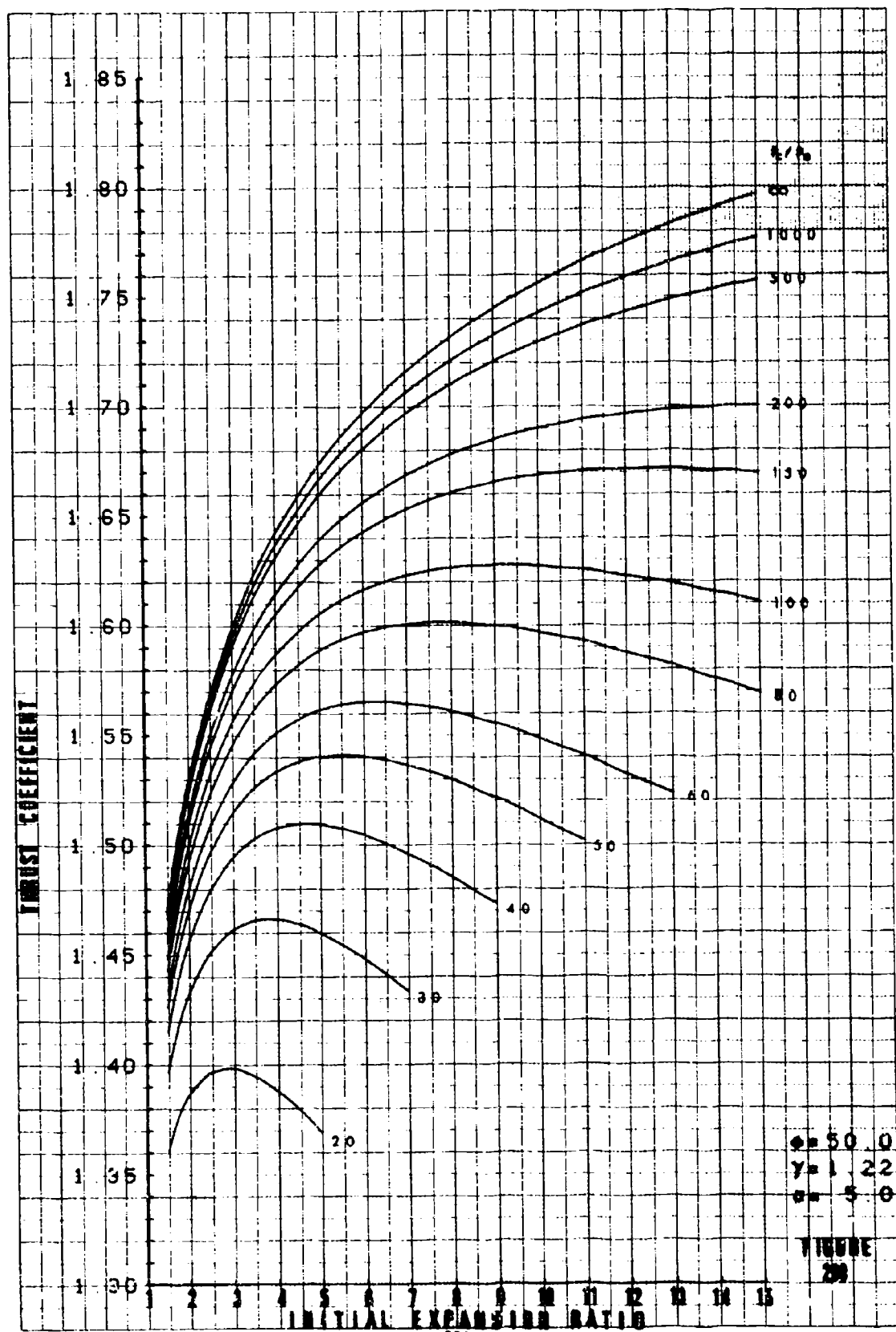


FIGURE
299

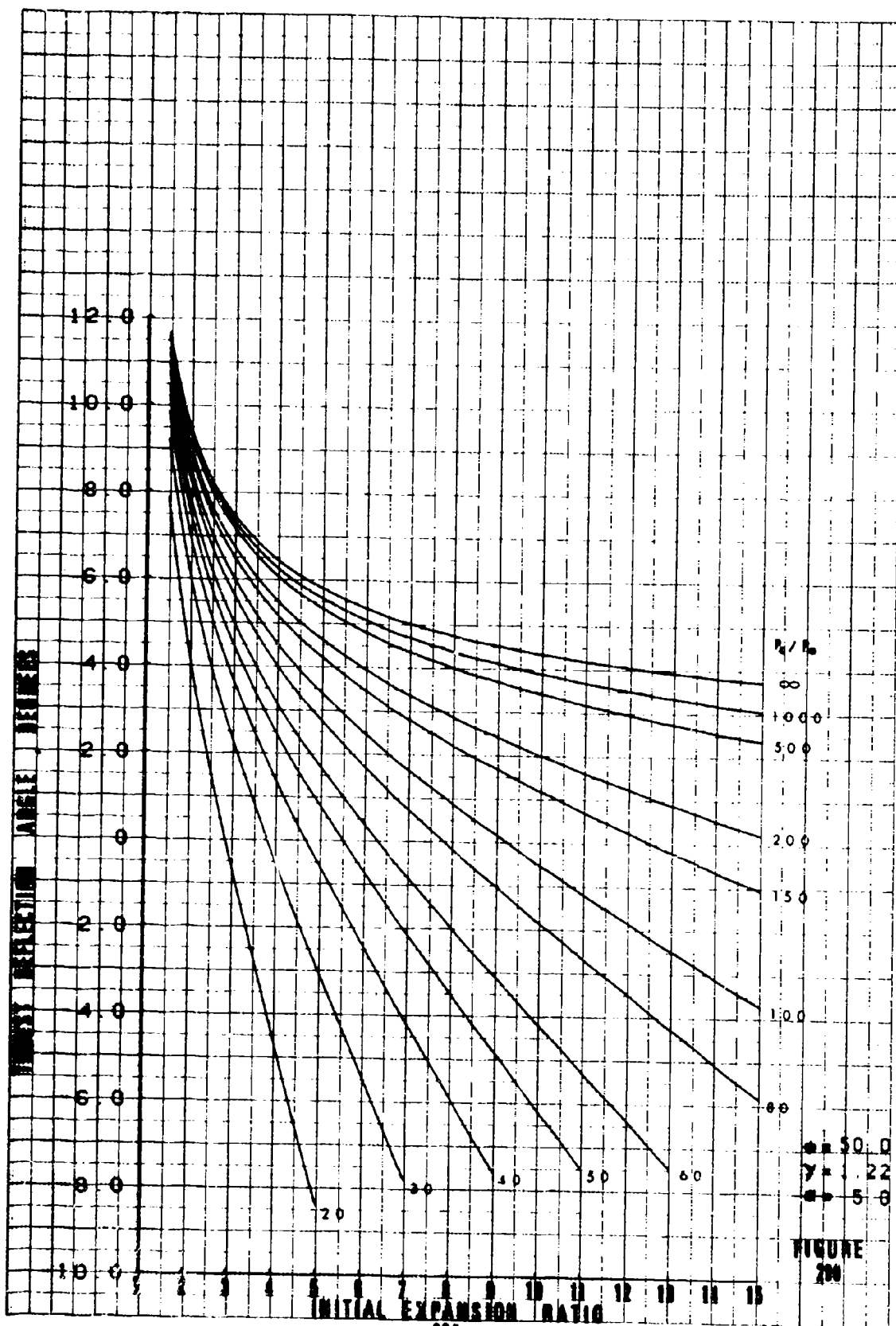
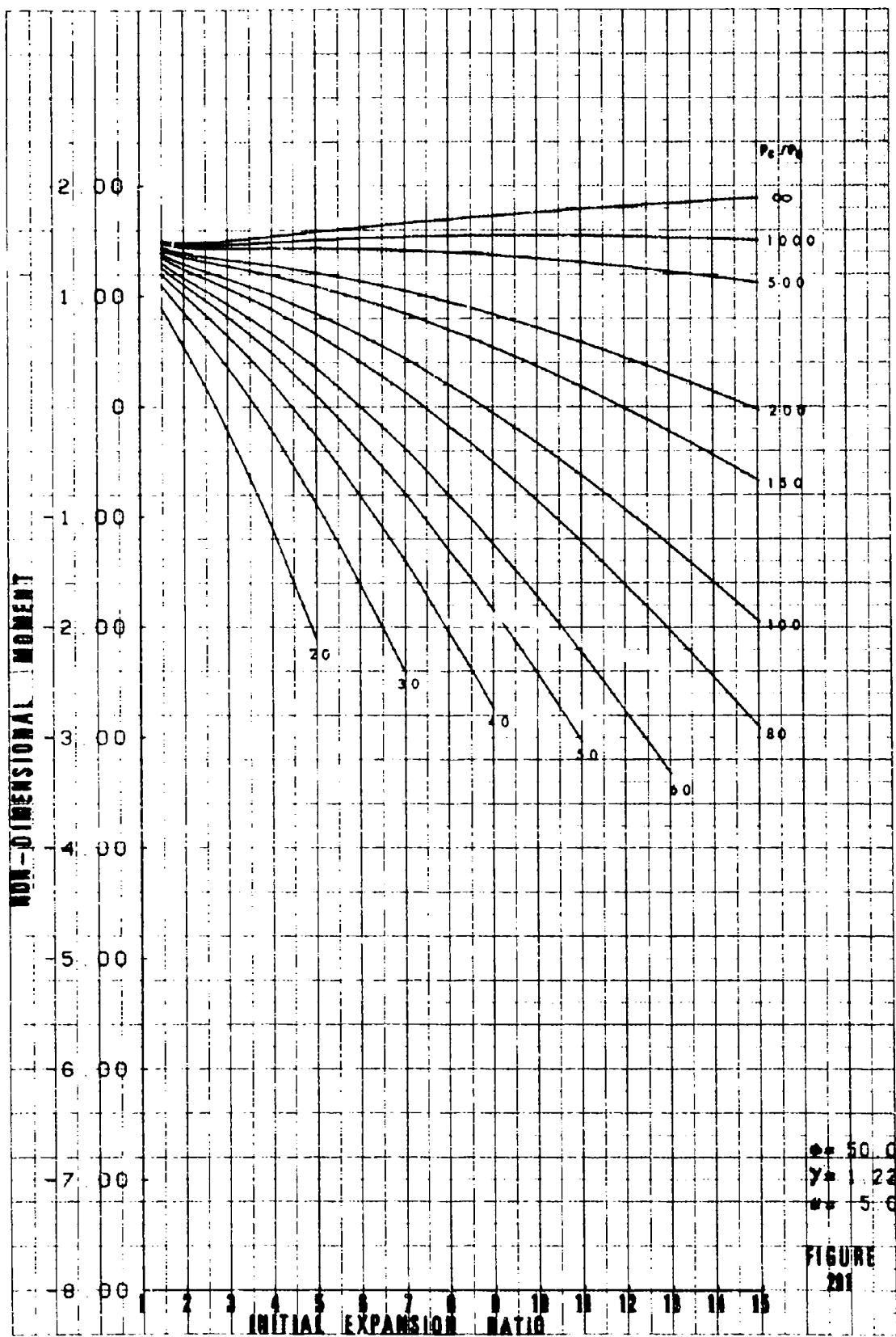
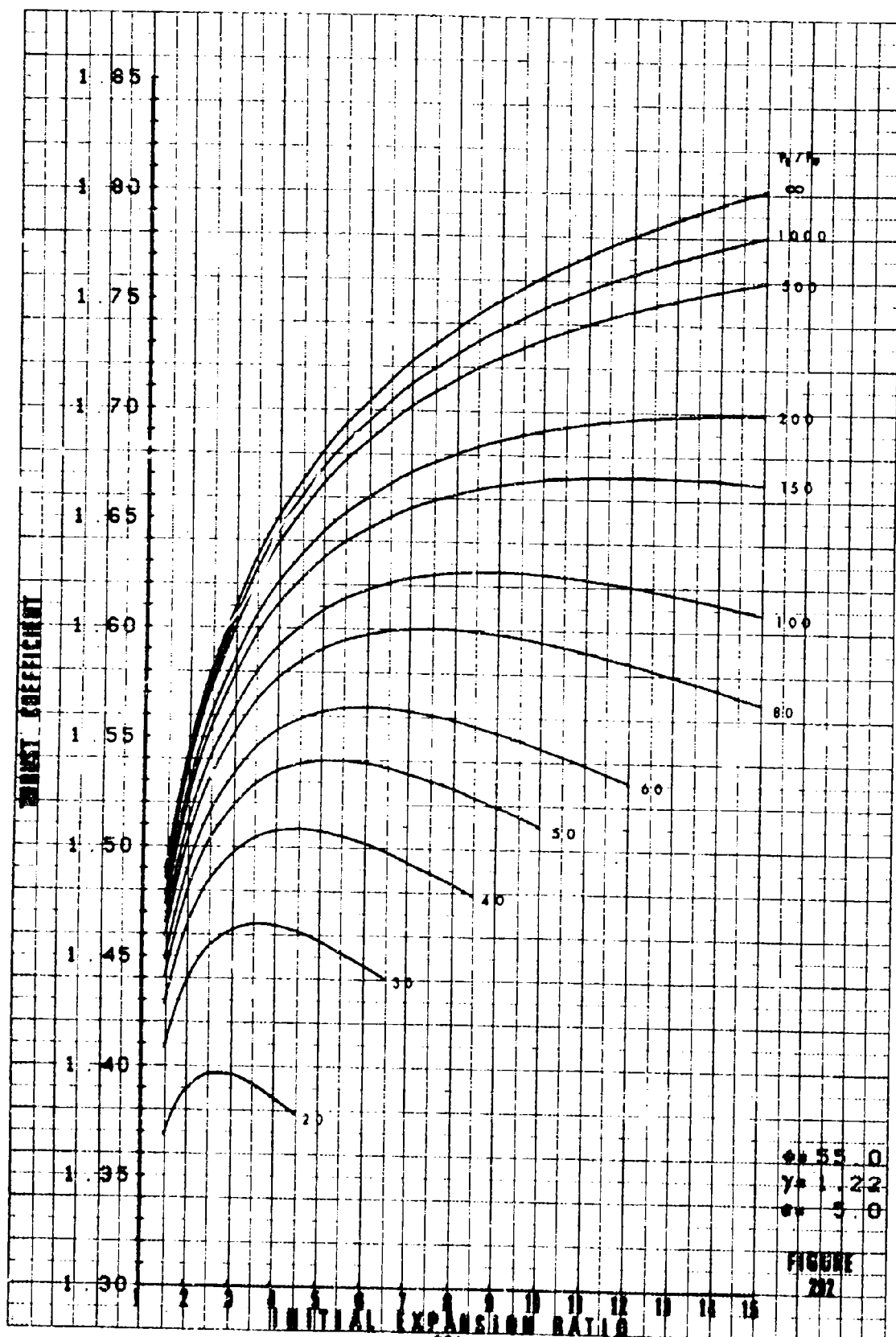


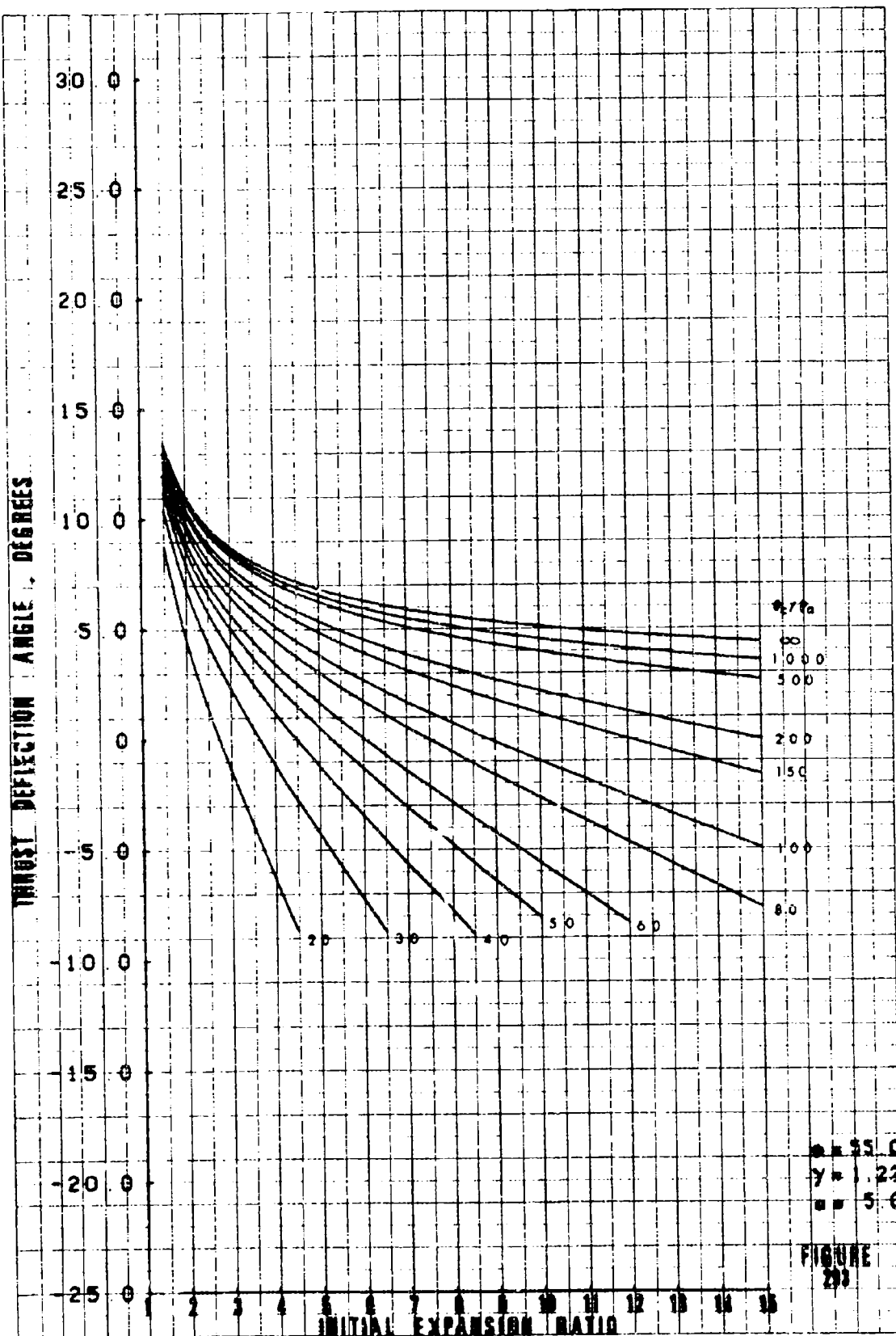
FIGURE 200





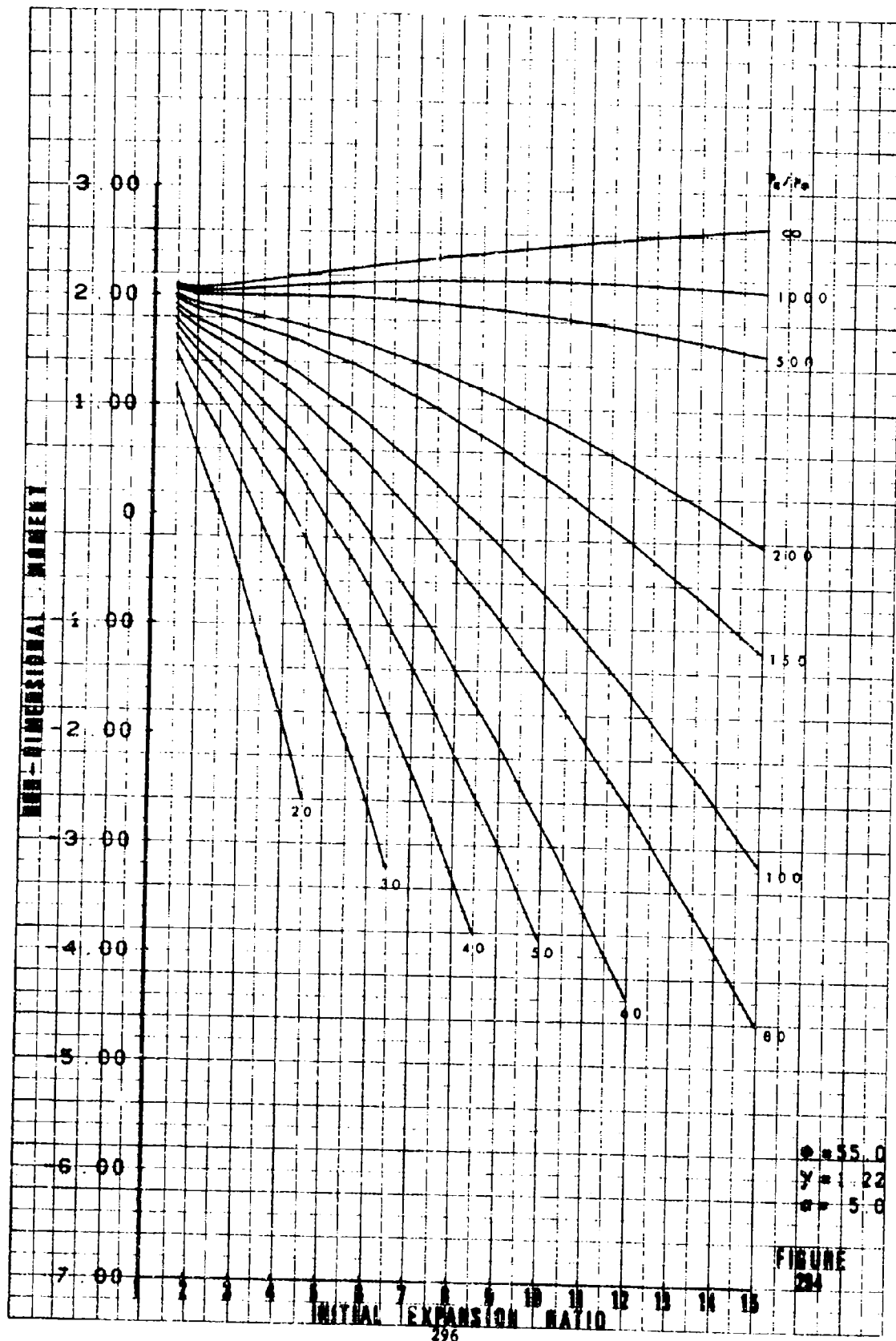
$\phi = 55.0$
 $\gamma = 1.22$
 $\sigma = 5.0$

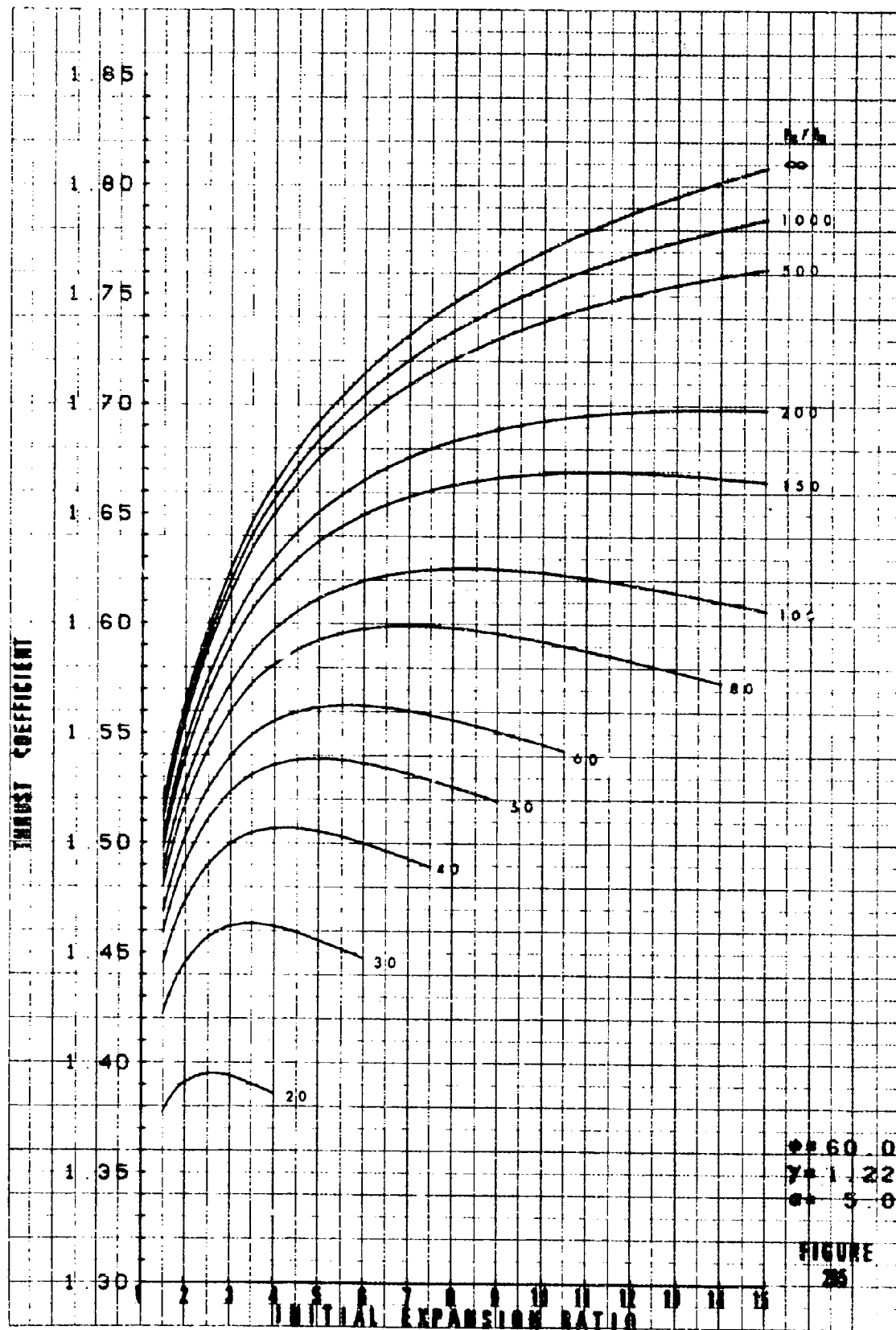
FIGURE 292



$\gamma = 1.22$
 $\gamma = 5.0$

FIGURE 295





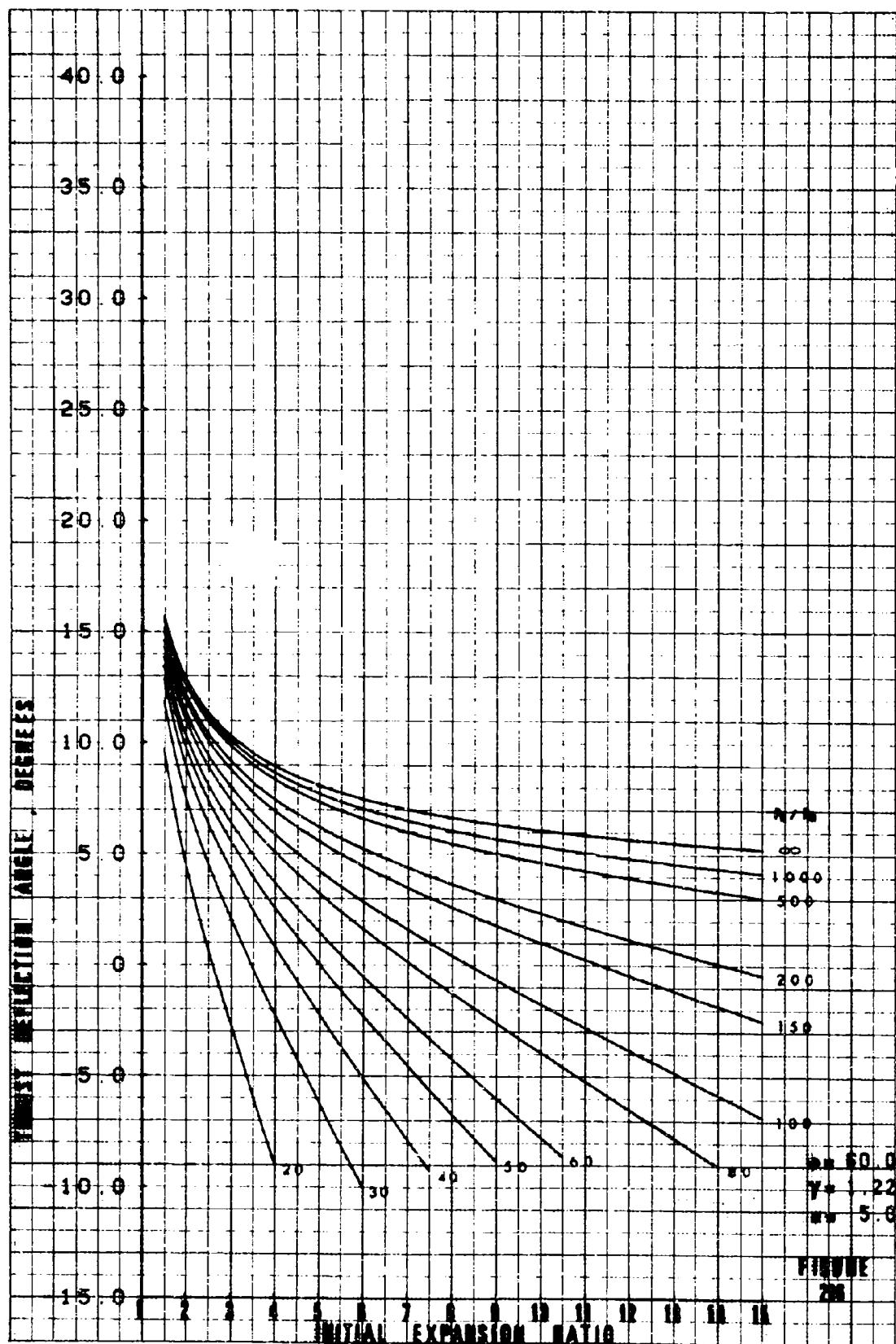
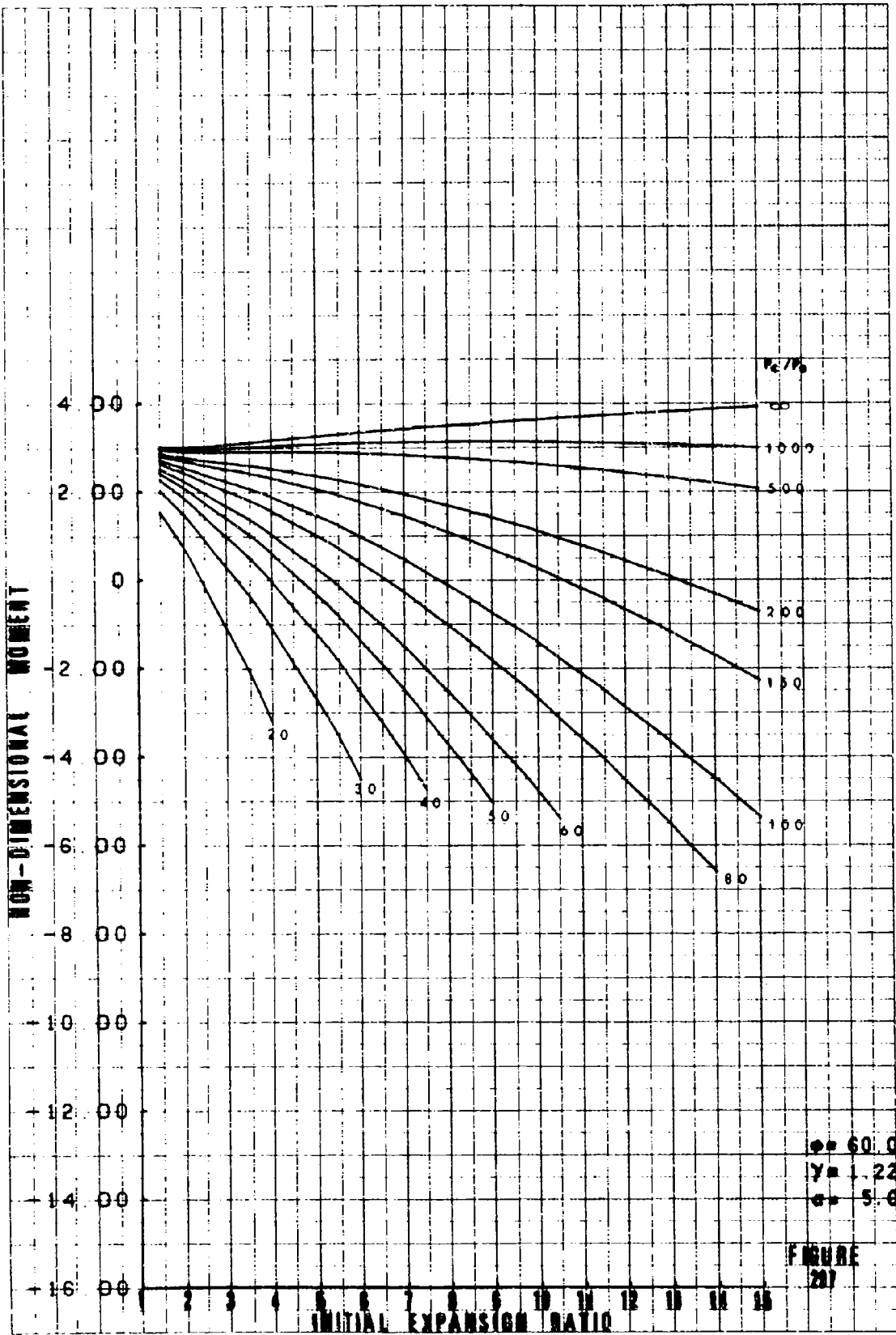
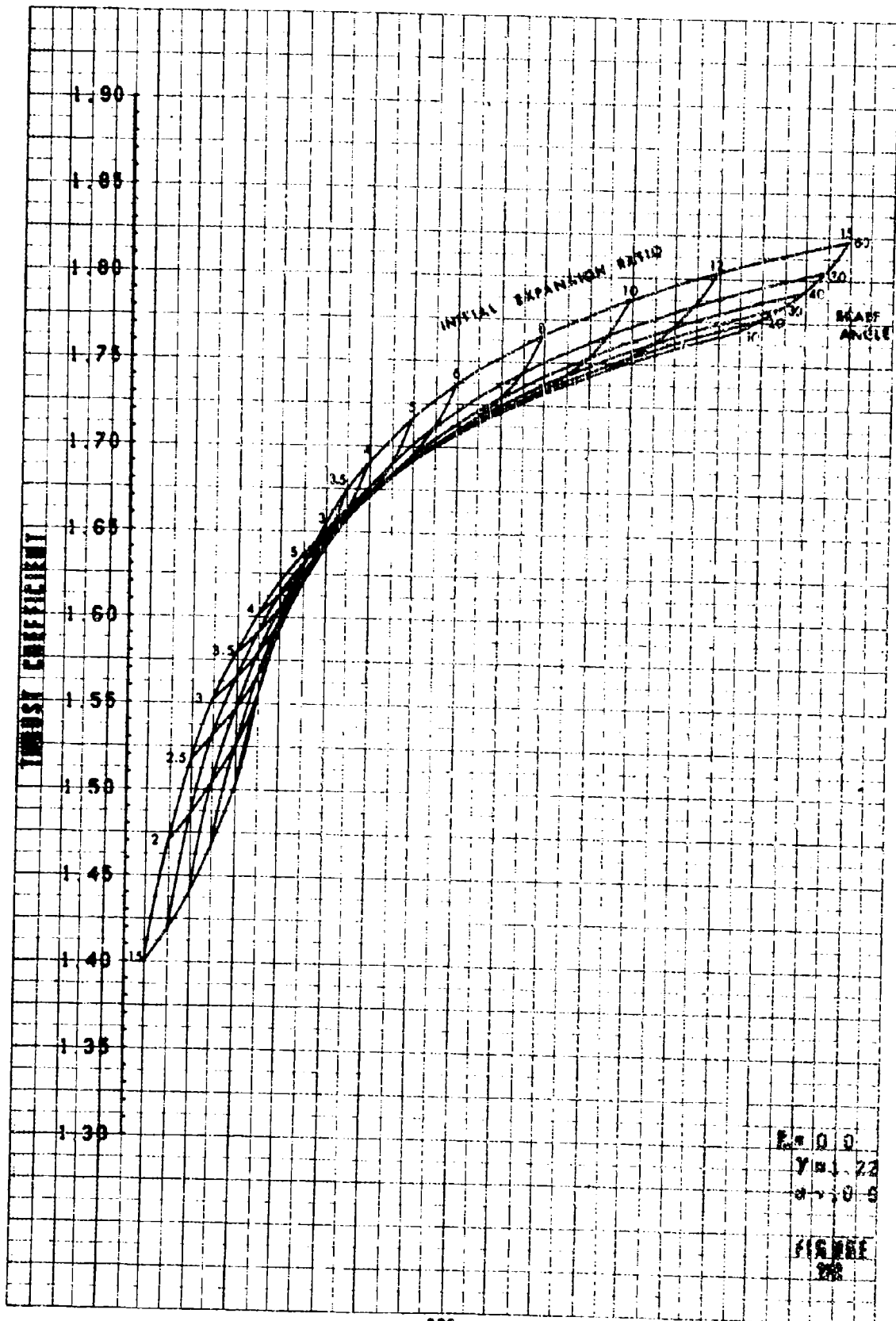


FIGURE 200



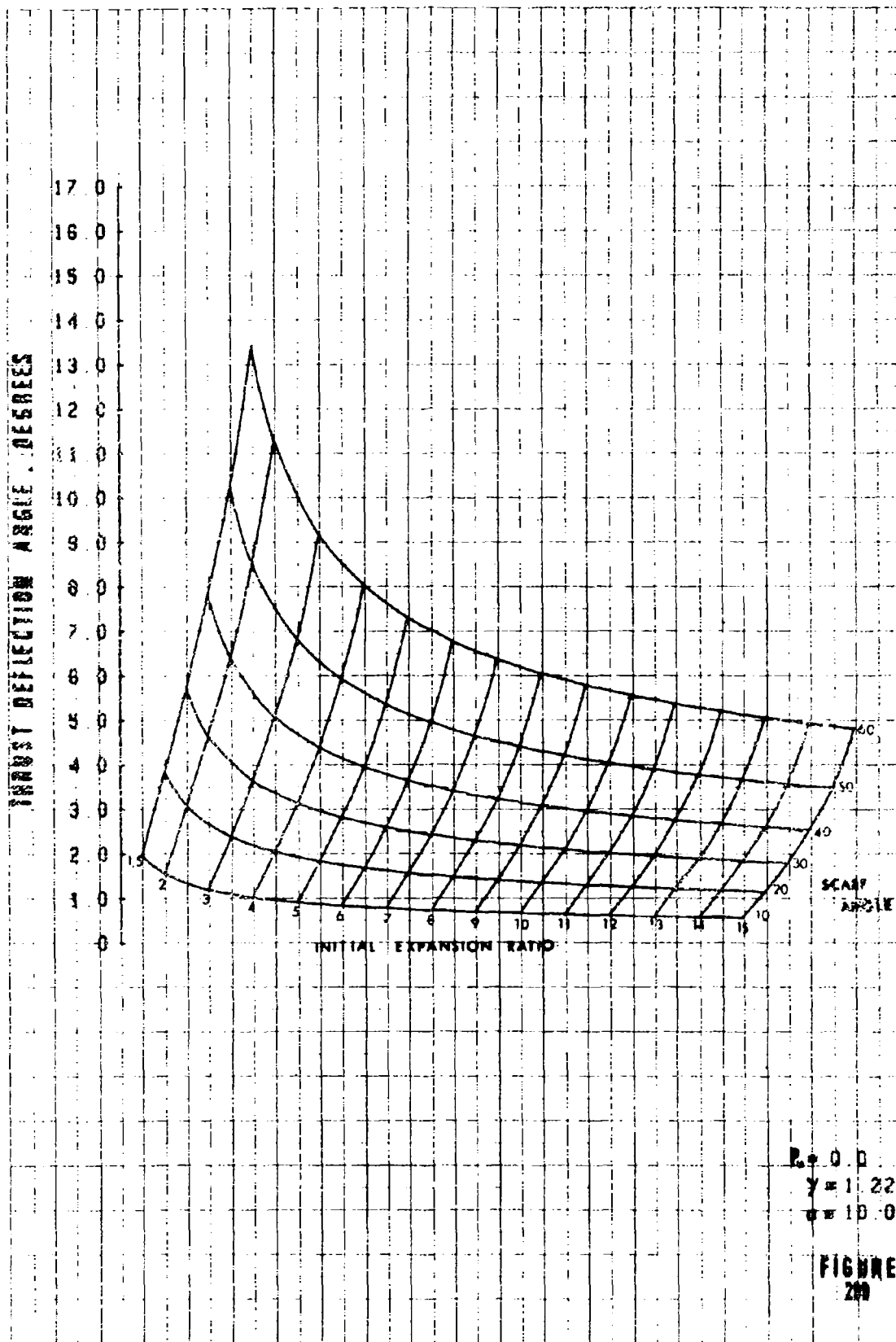
$\bullet = 60.0$
 $\gamma = 1.22$
 $\sigma = 5.6$

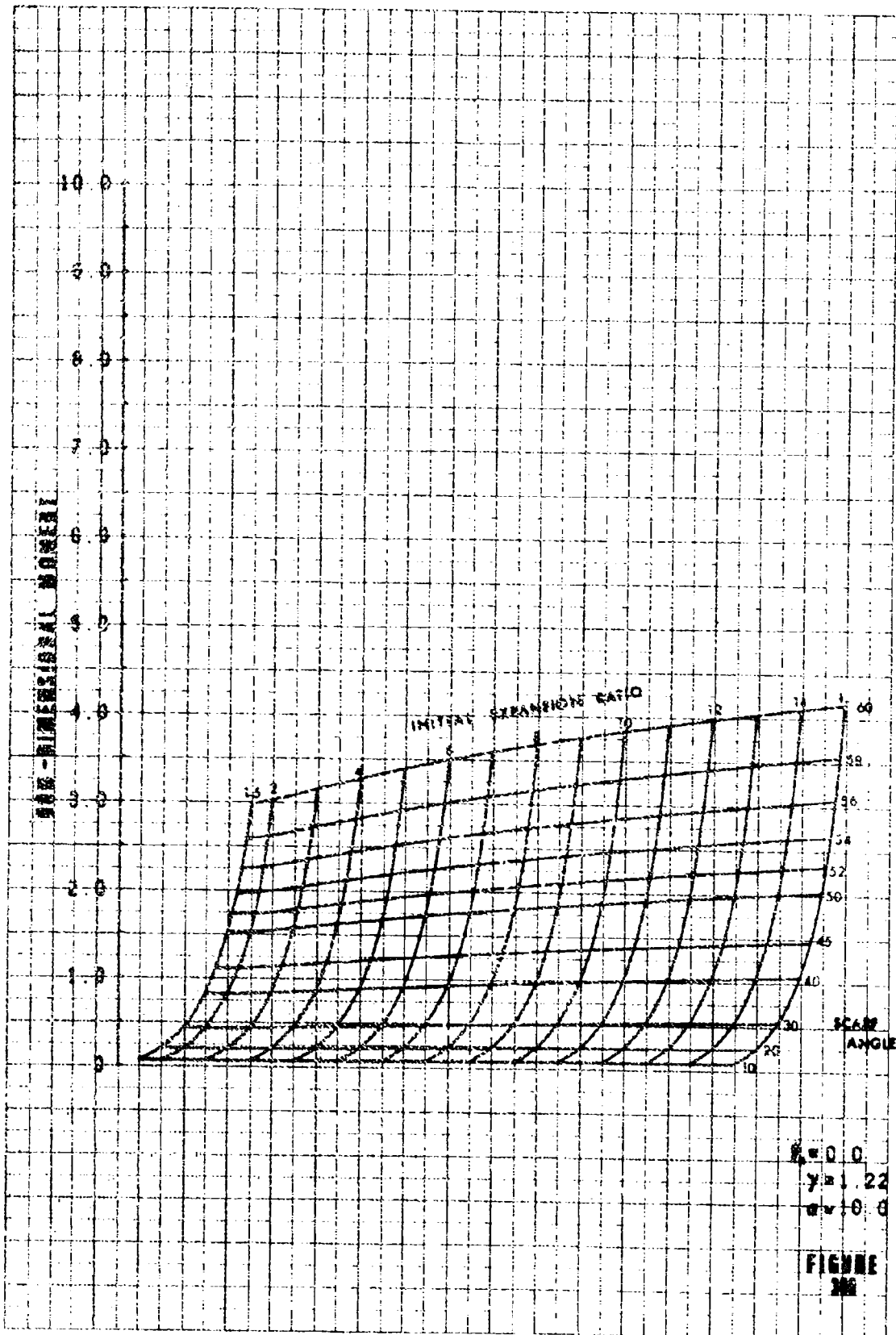
FIGURE 207

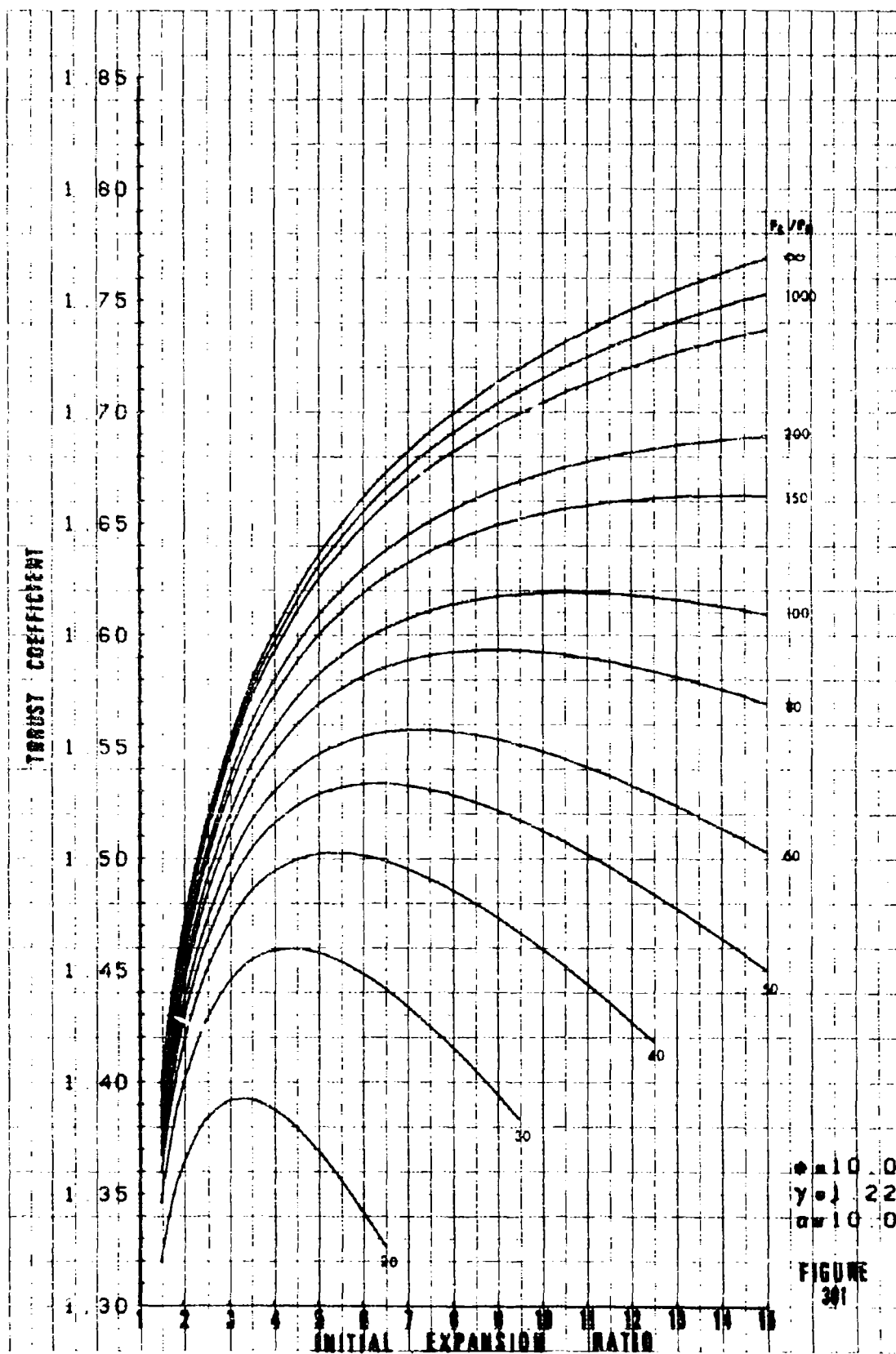


$E = 0.0$
 $\gamma = 1.22$
 $\sigma = 0.0$

FIGURE
 28







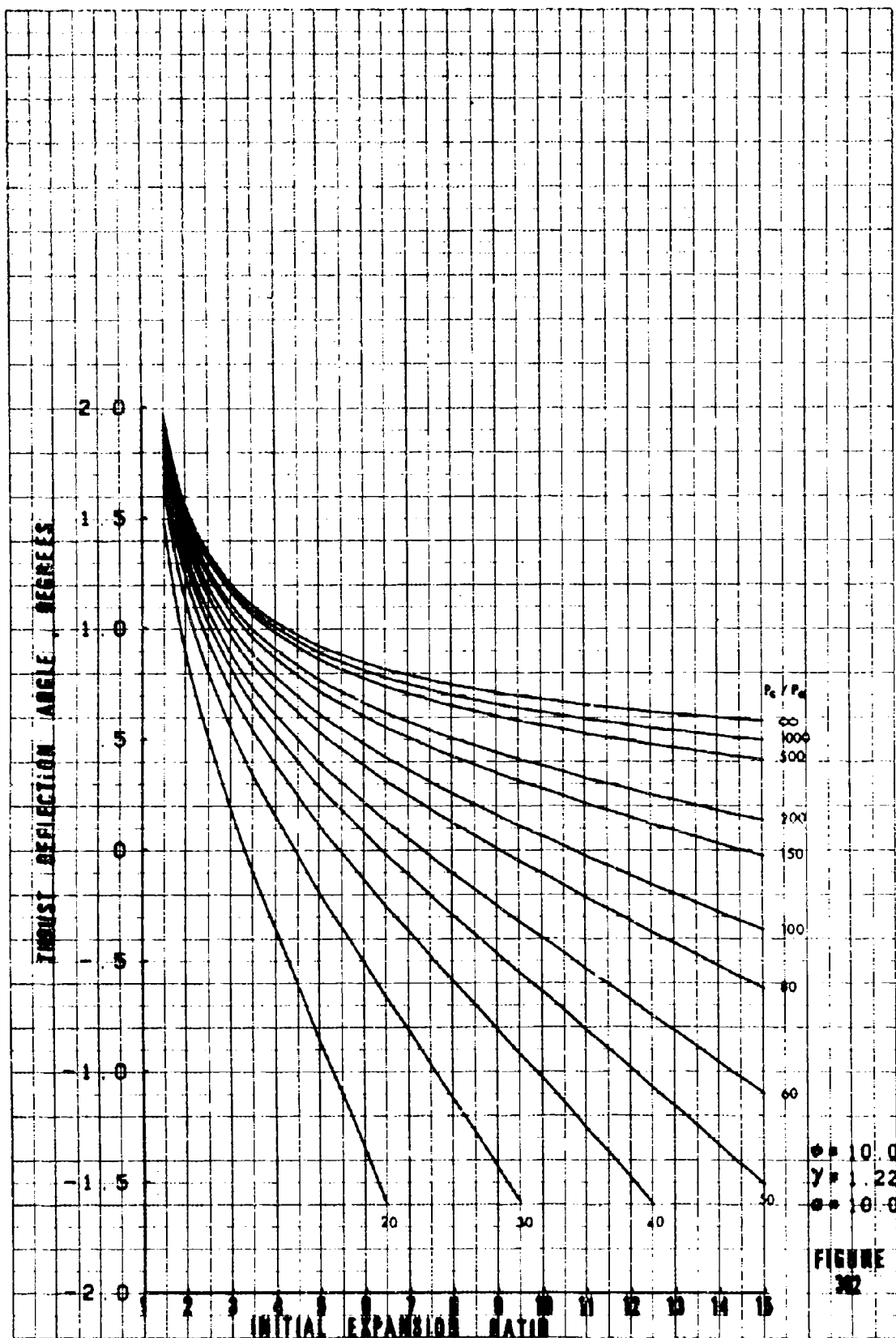
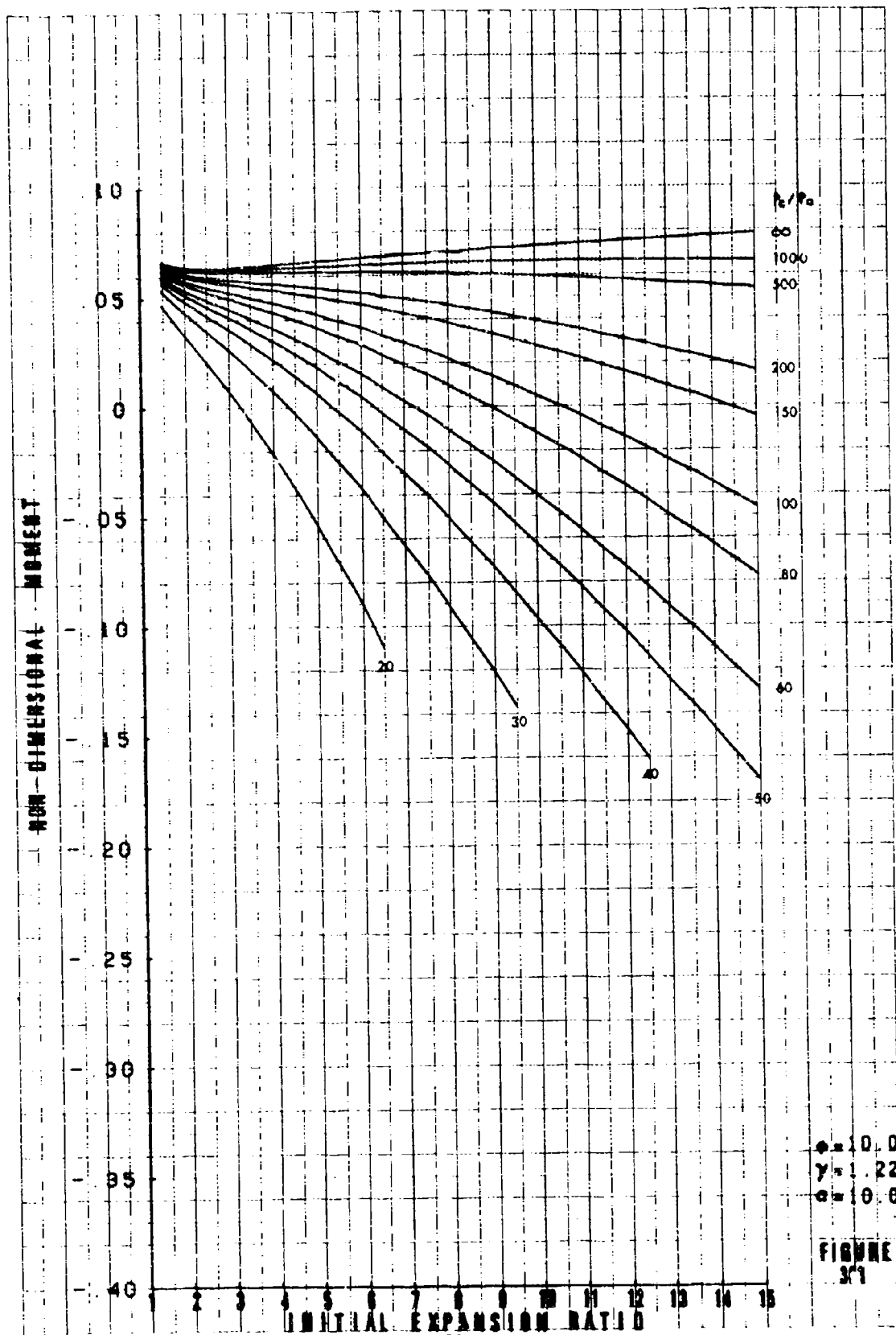
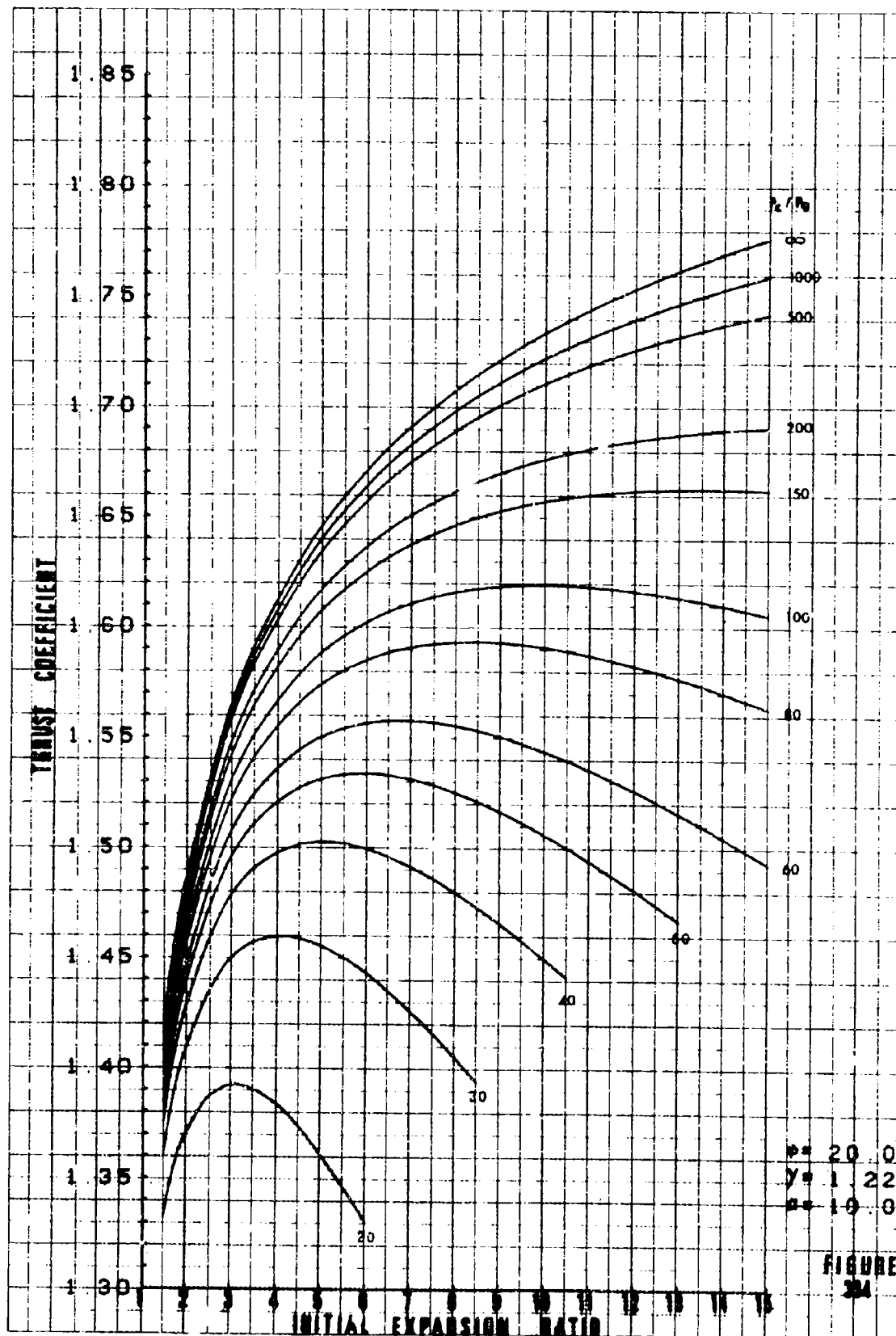
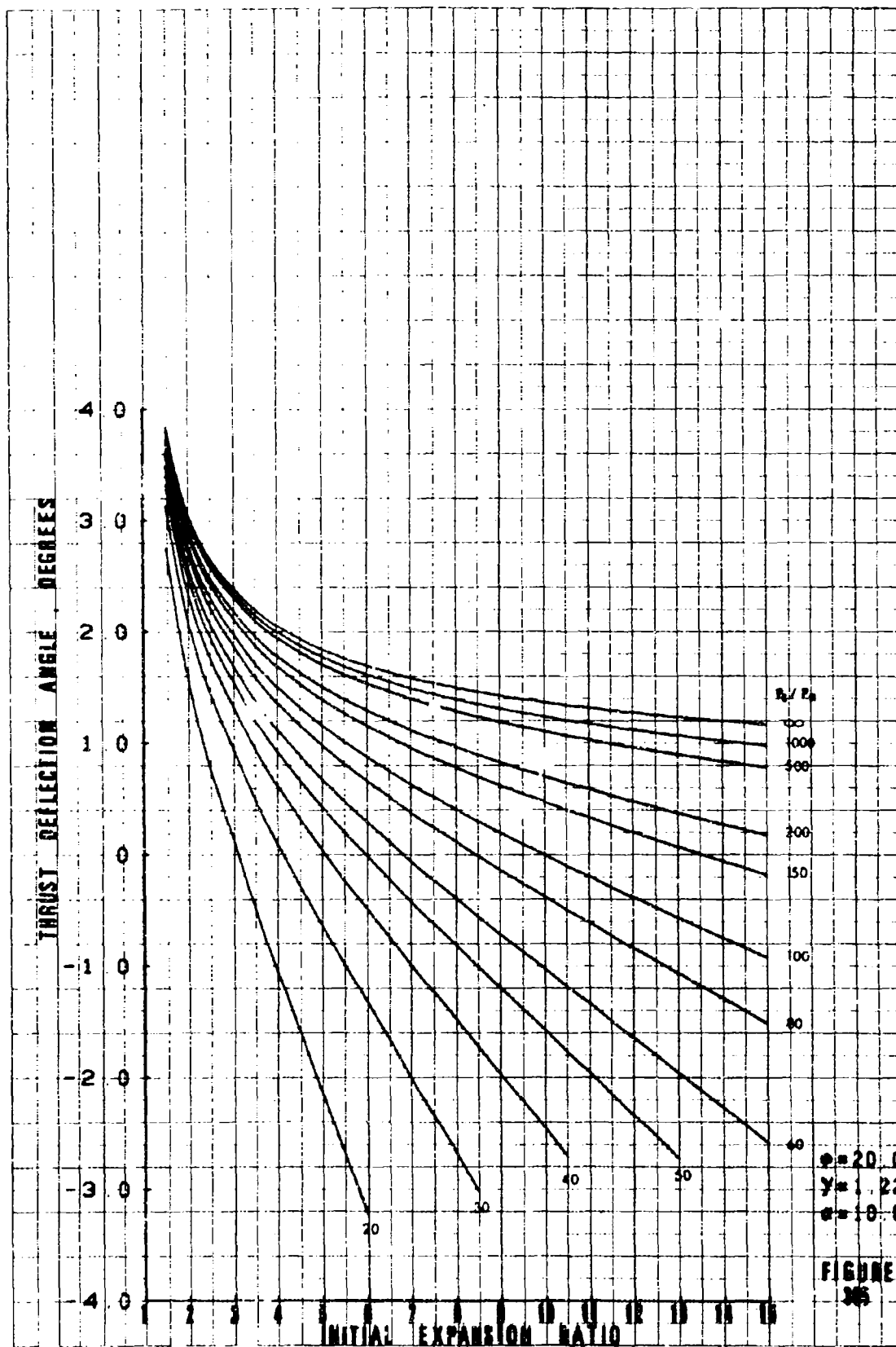


FIGURE 302

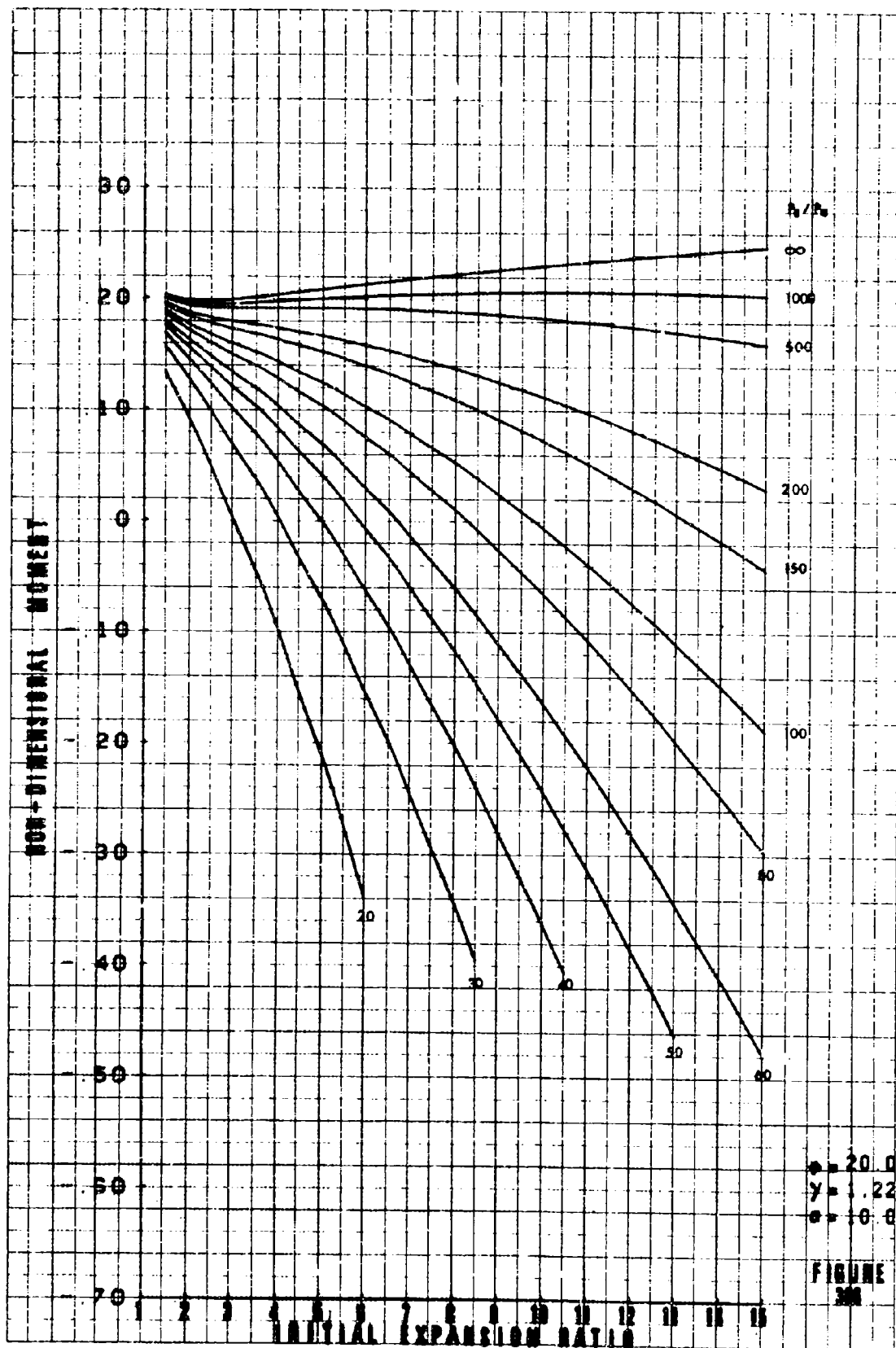






$\gamma = 20.0$
 $\gamma = 1.22$
 $\sigma = 10.0$

FIGURE 36



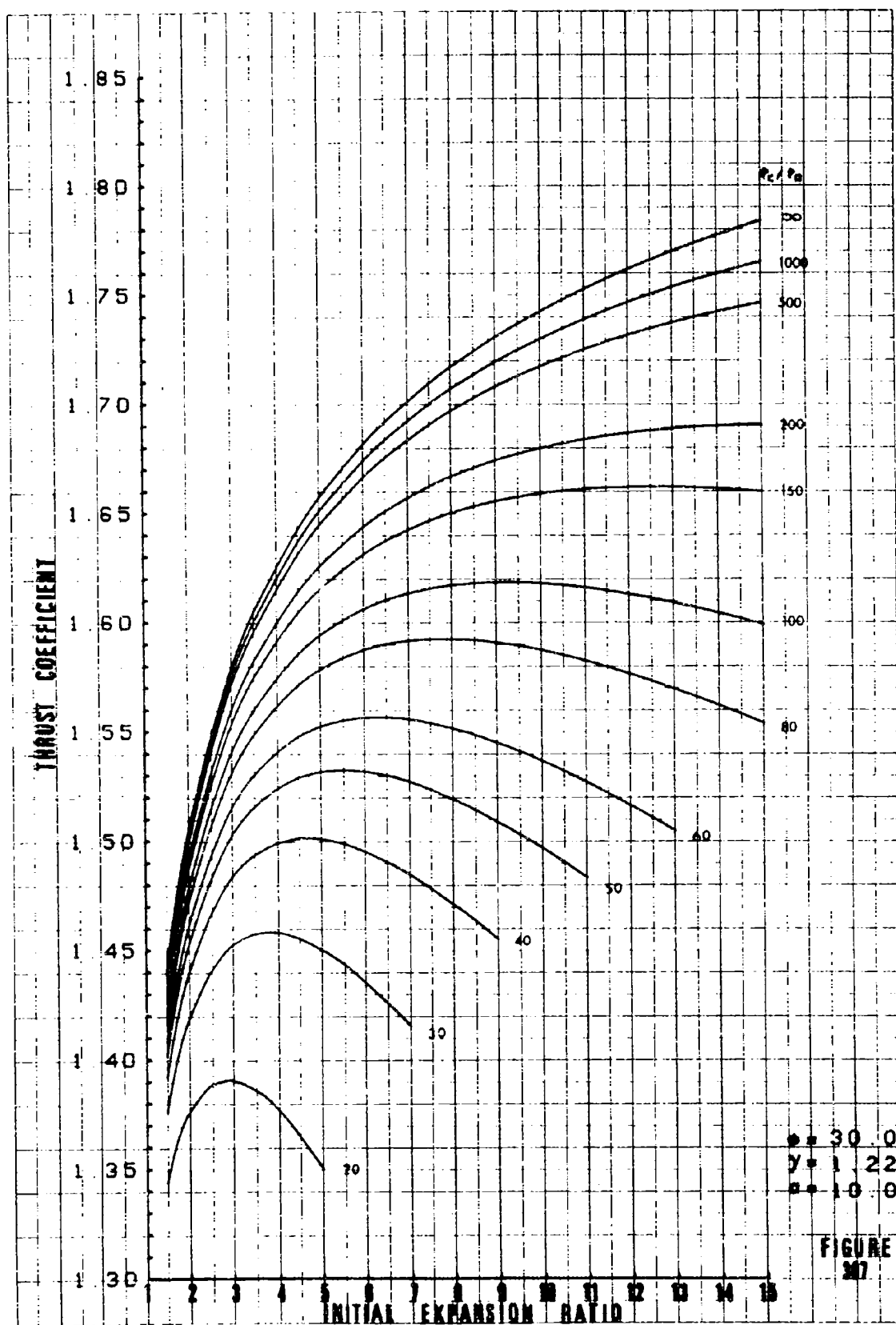
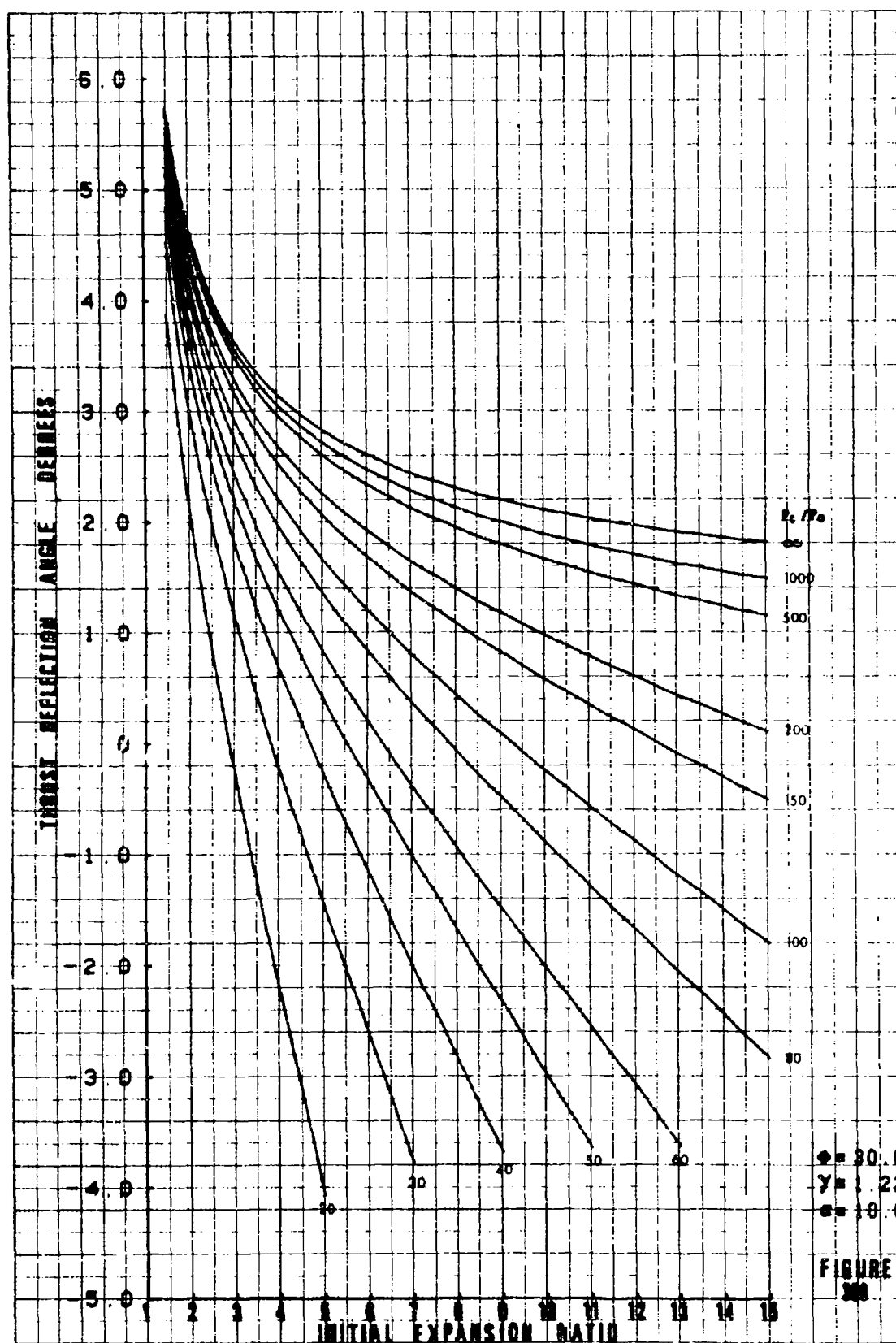
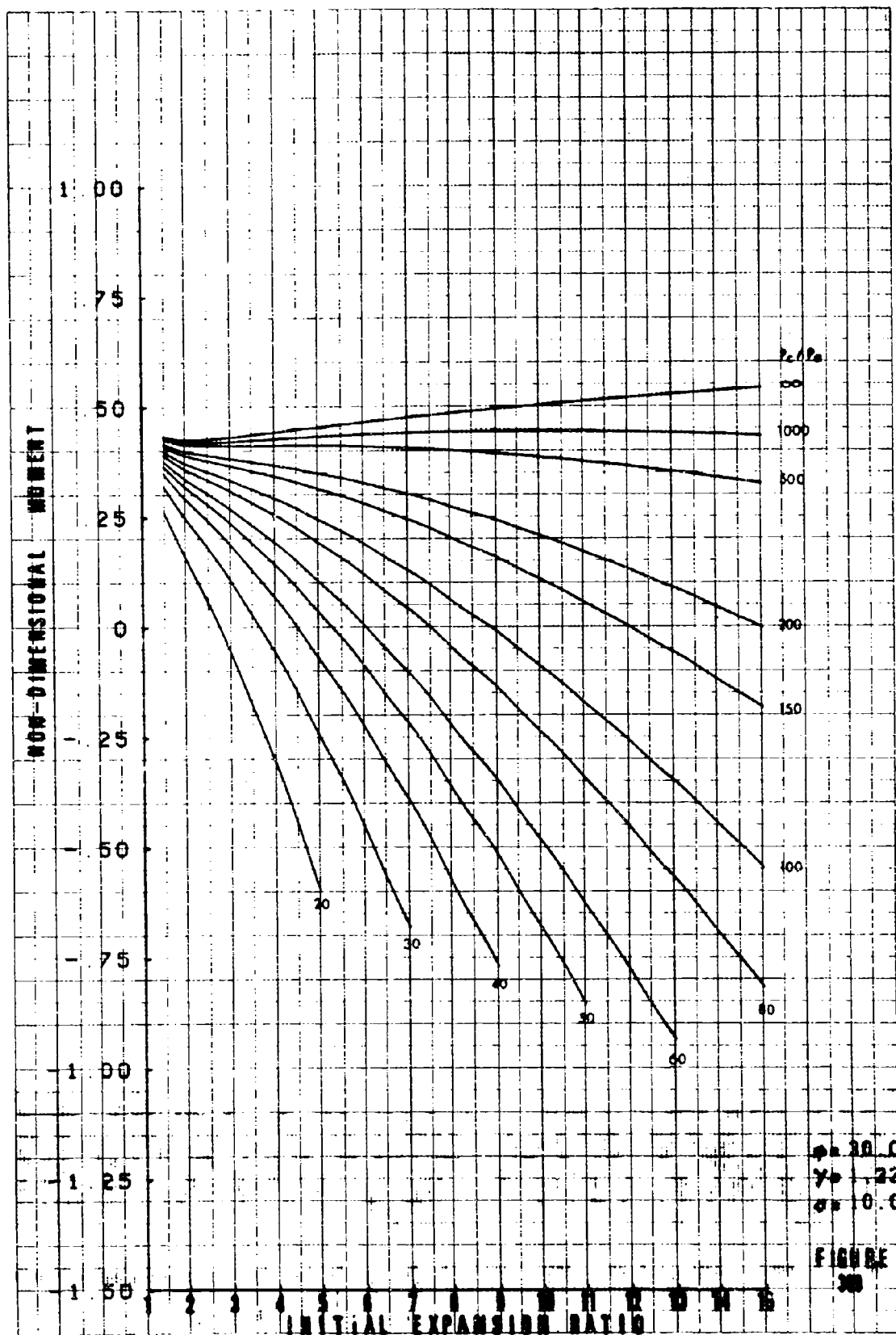


FIGURE 207



$\phi = 30.0$
 $\gamma = 1.22$
 $\sigma = 10.6$

FIGURE 200



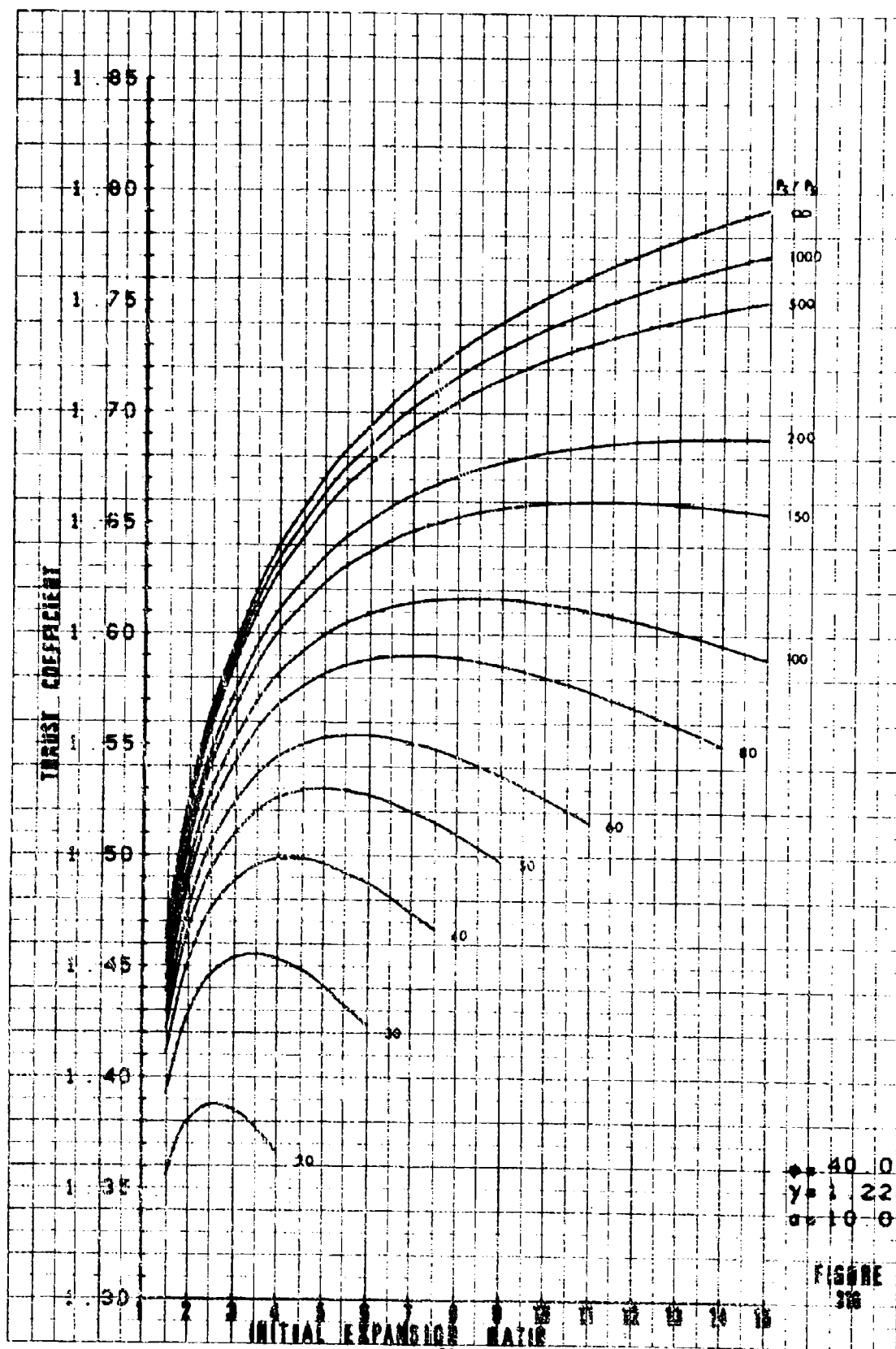


FIGURE 308

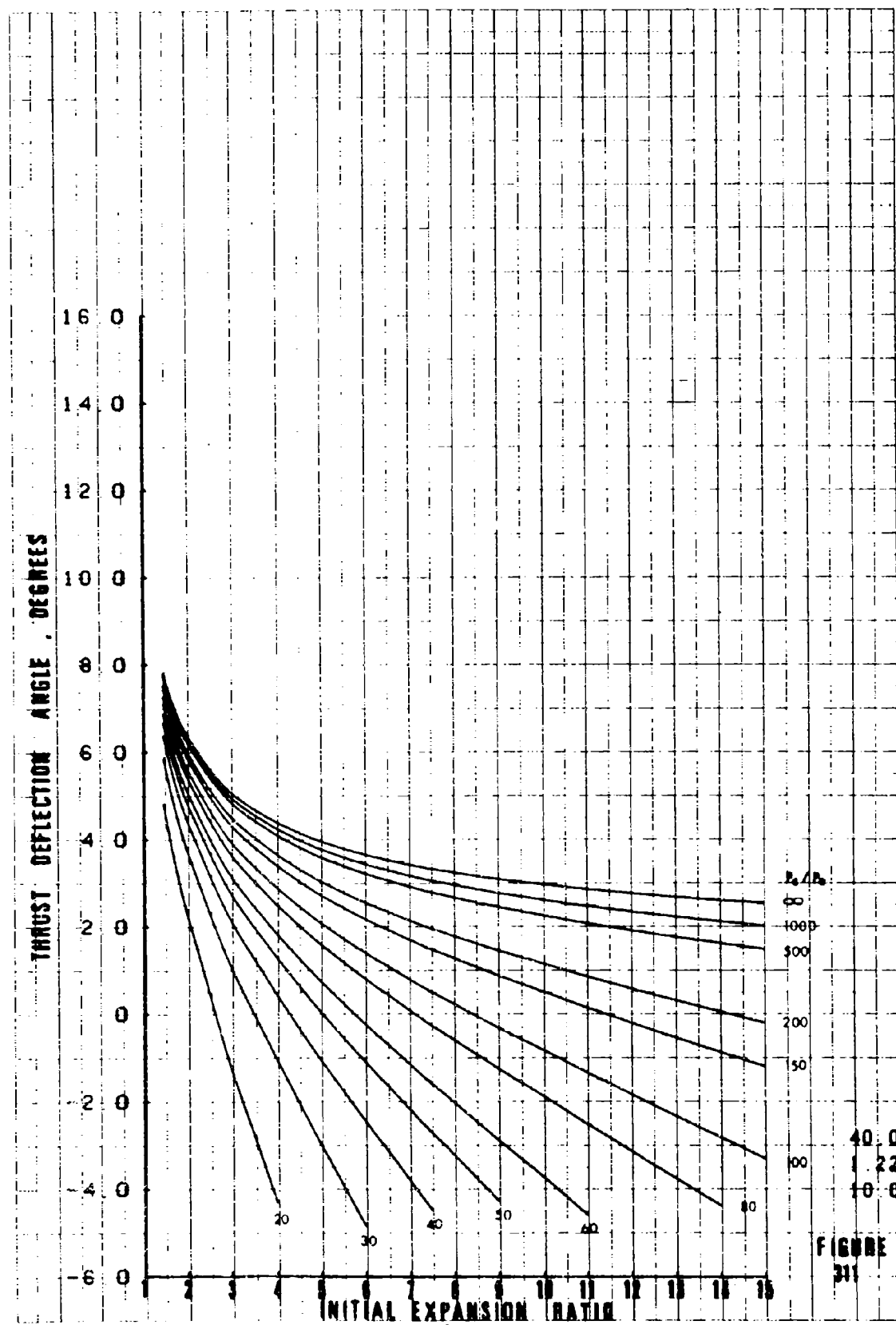
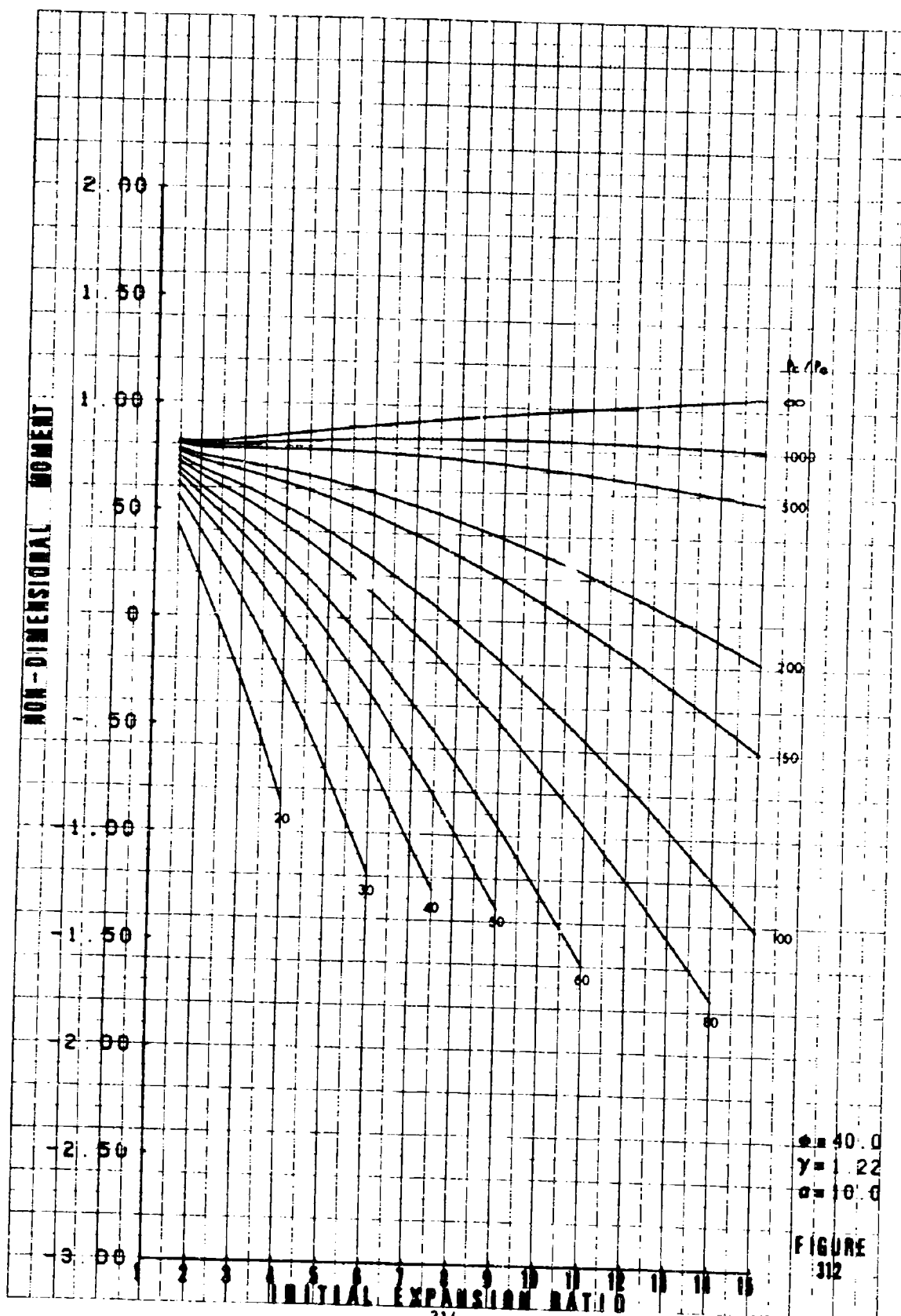


FIGURE 311



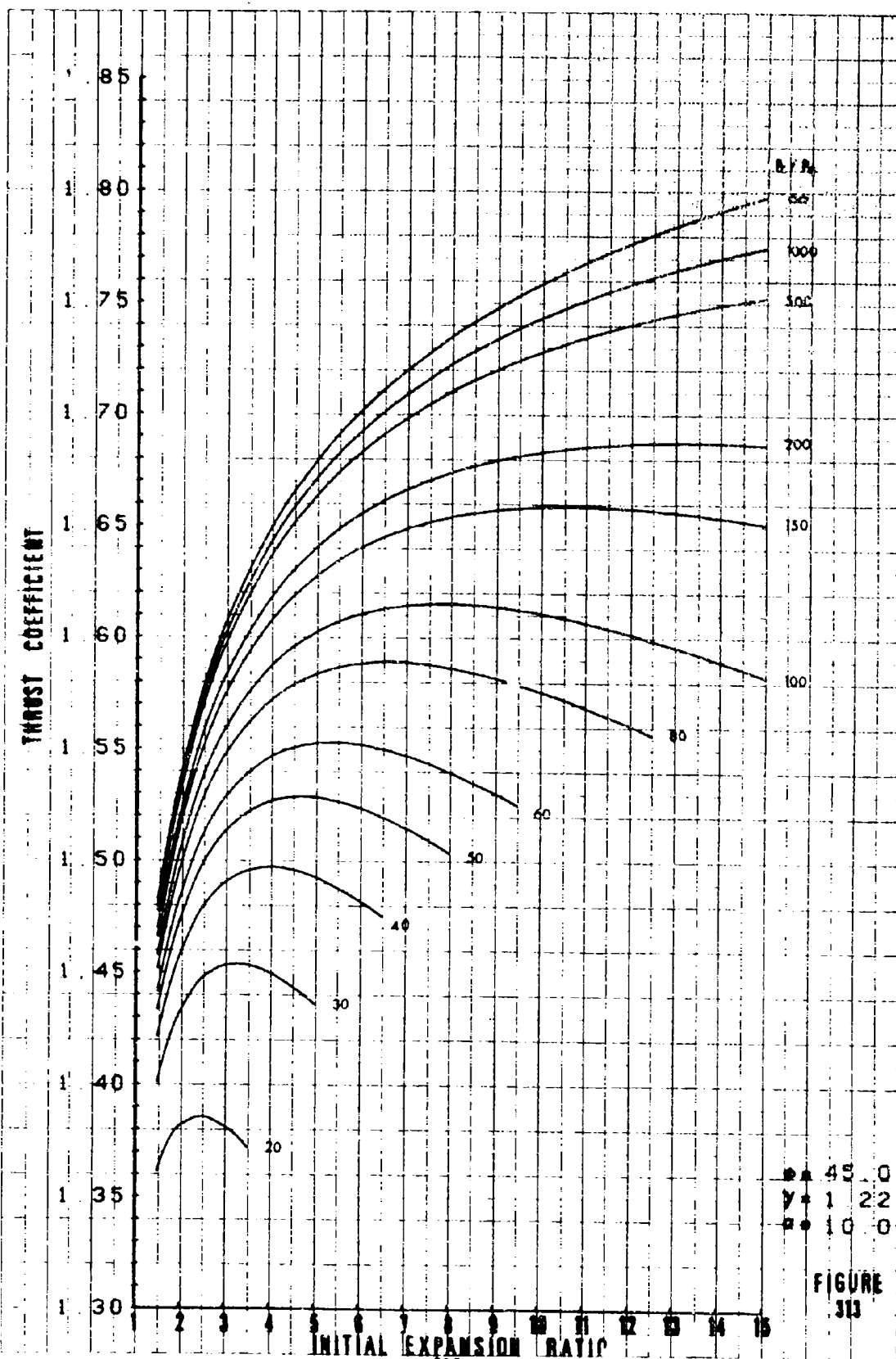
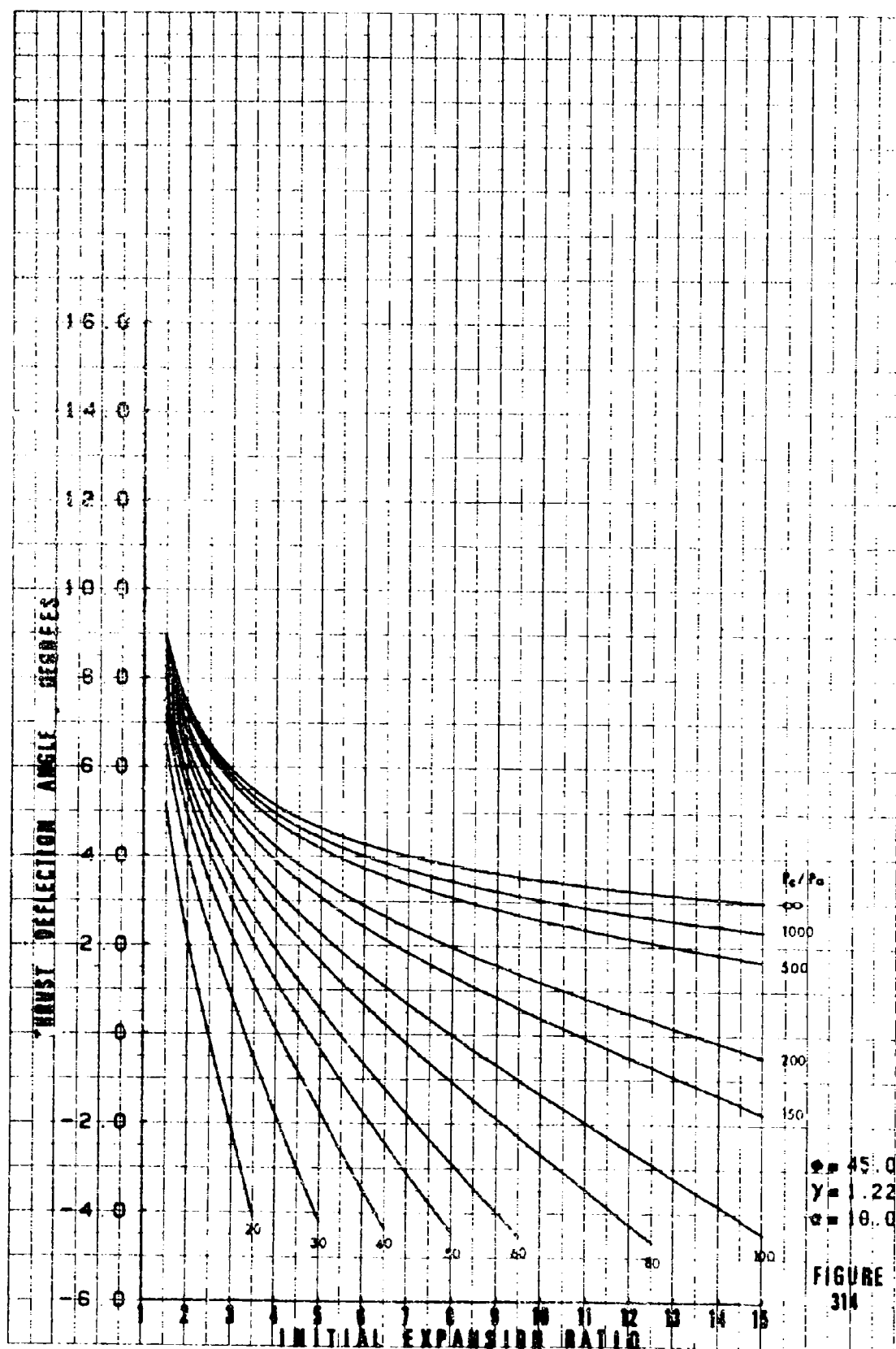
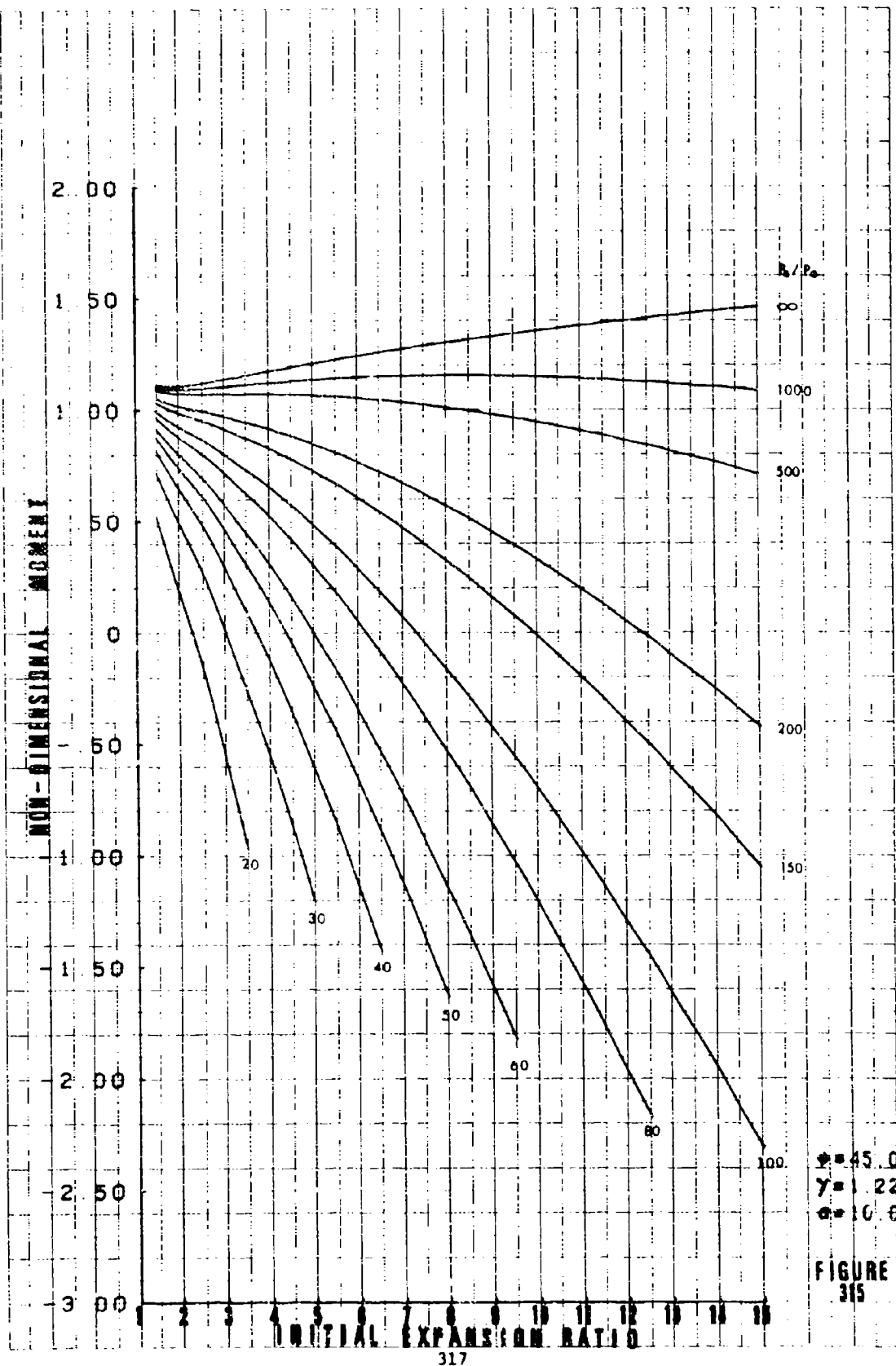


FIGURE 313



$\phi = 45.0$
 $\gamma = 1.22$
 $\sigma = 10.0$

FIGURE 314



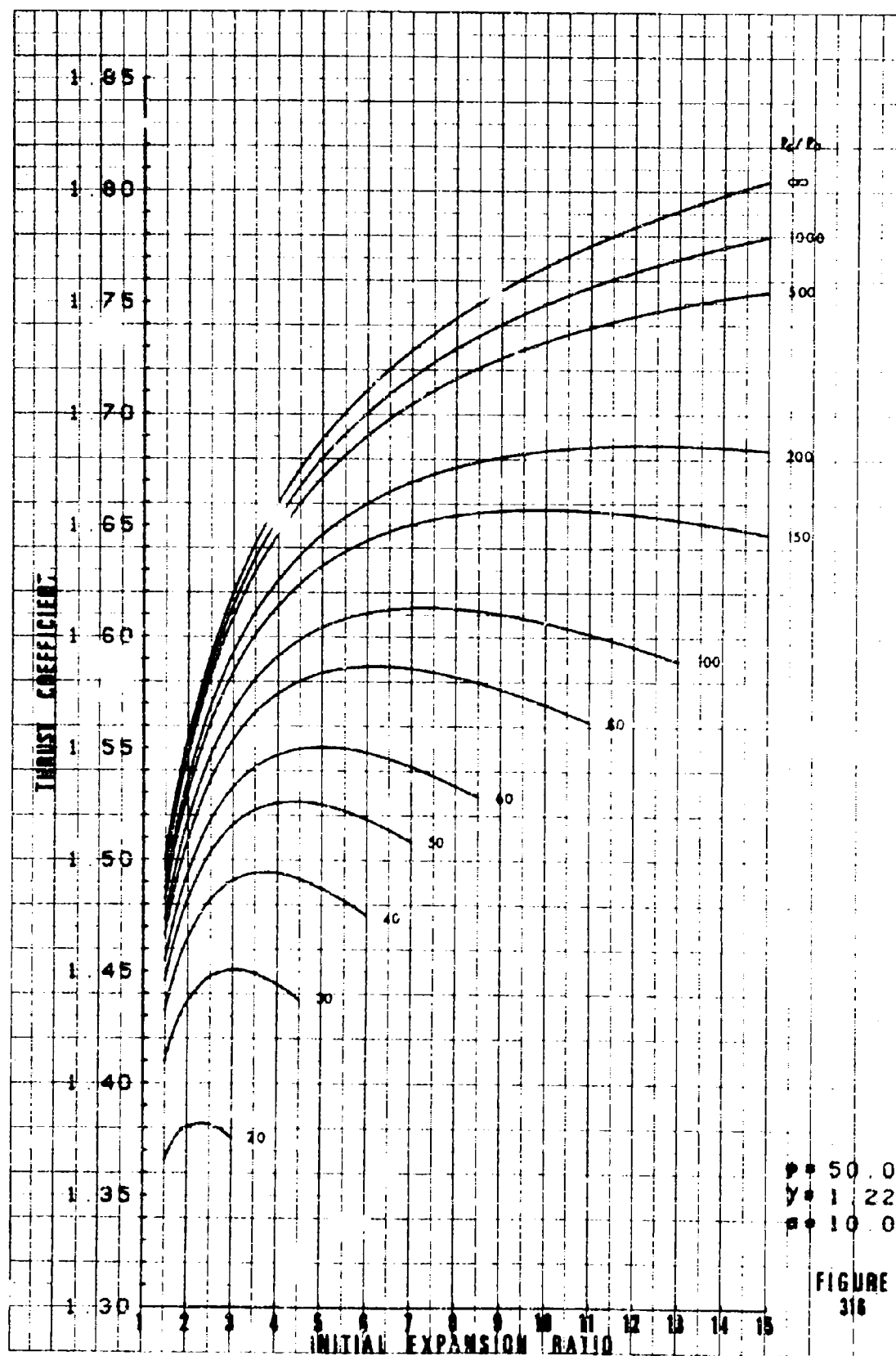


FIGURE
318

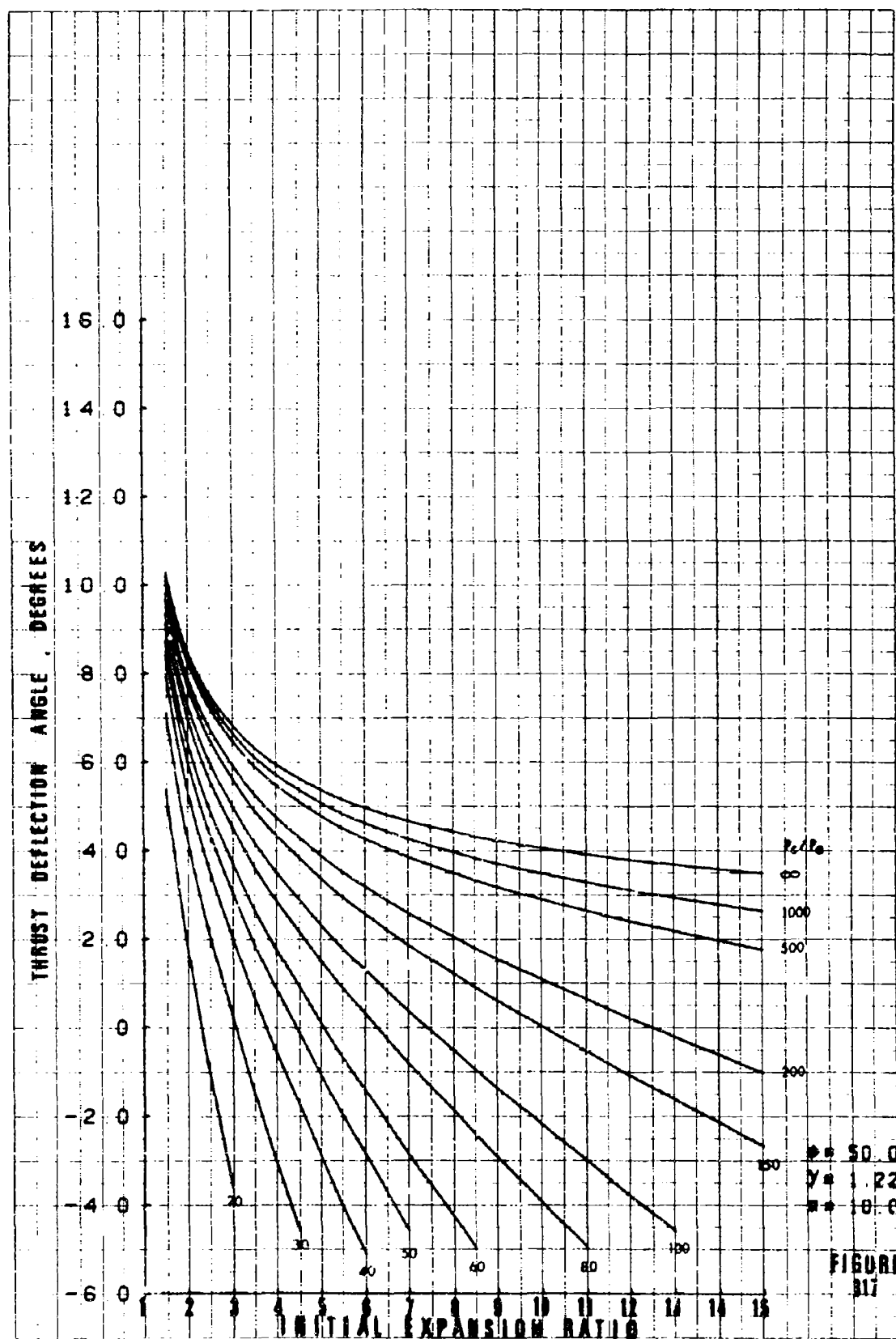
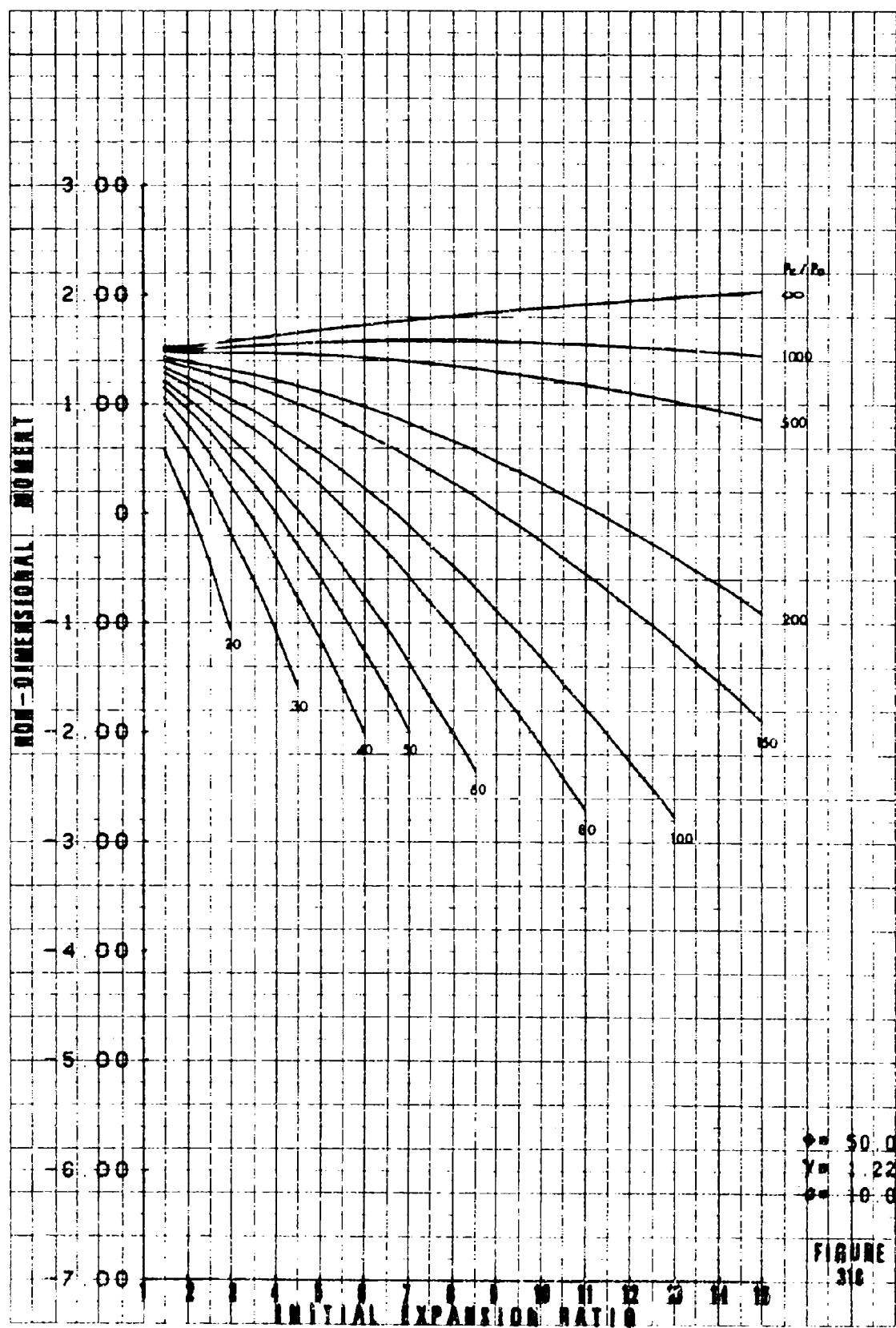


FIGURE 317



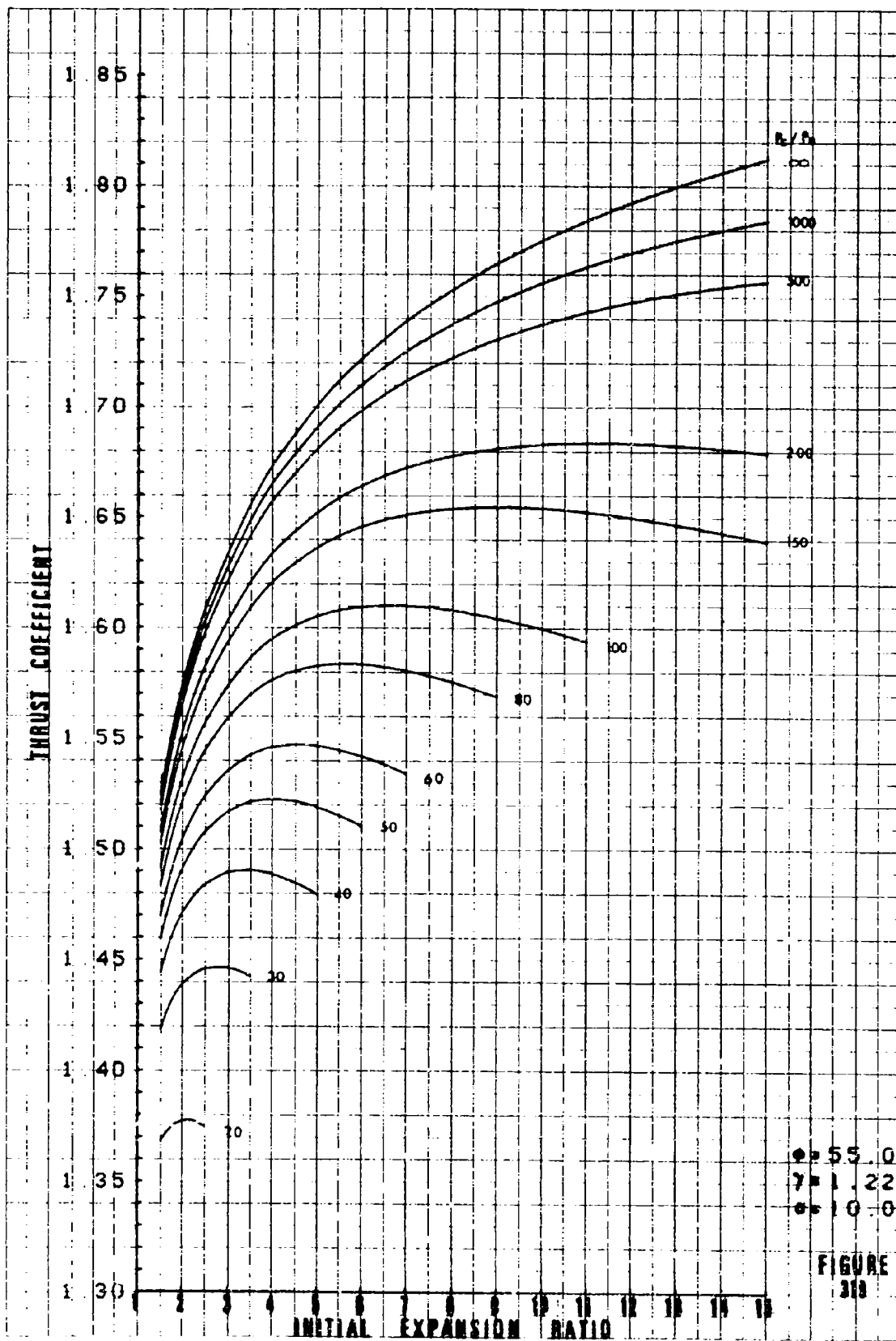
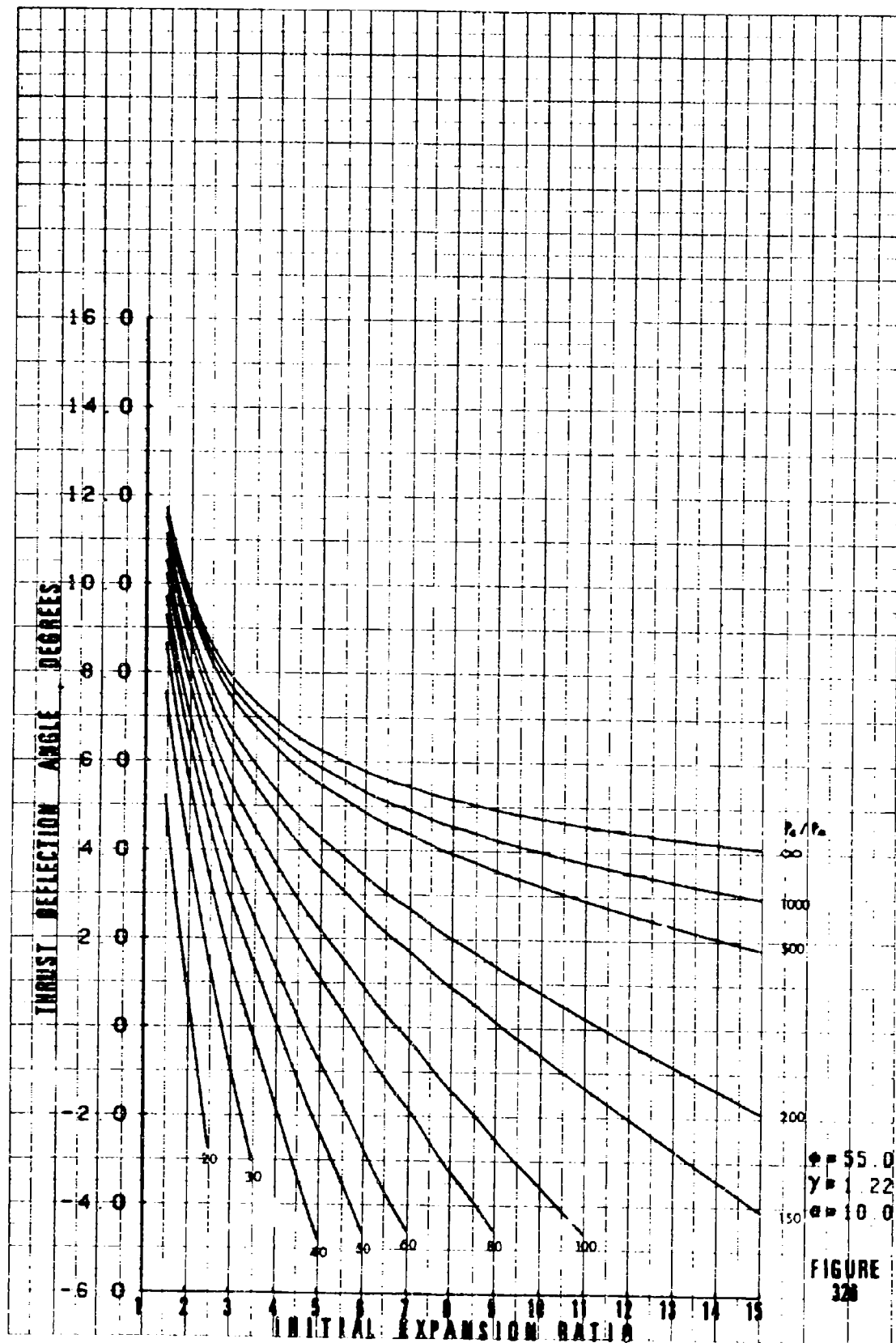


FIGURE 319



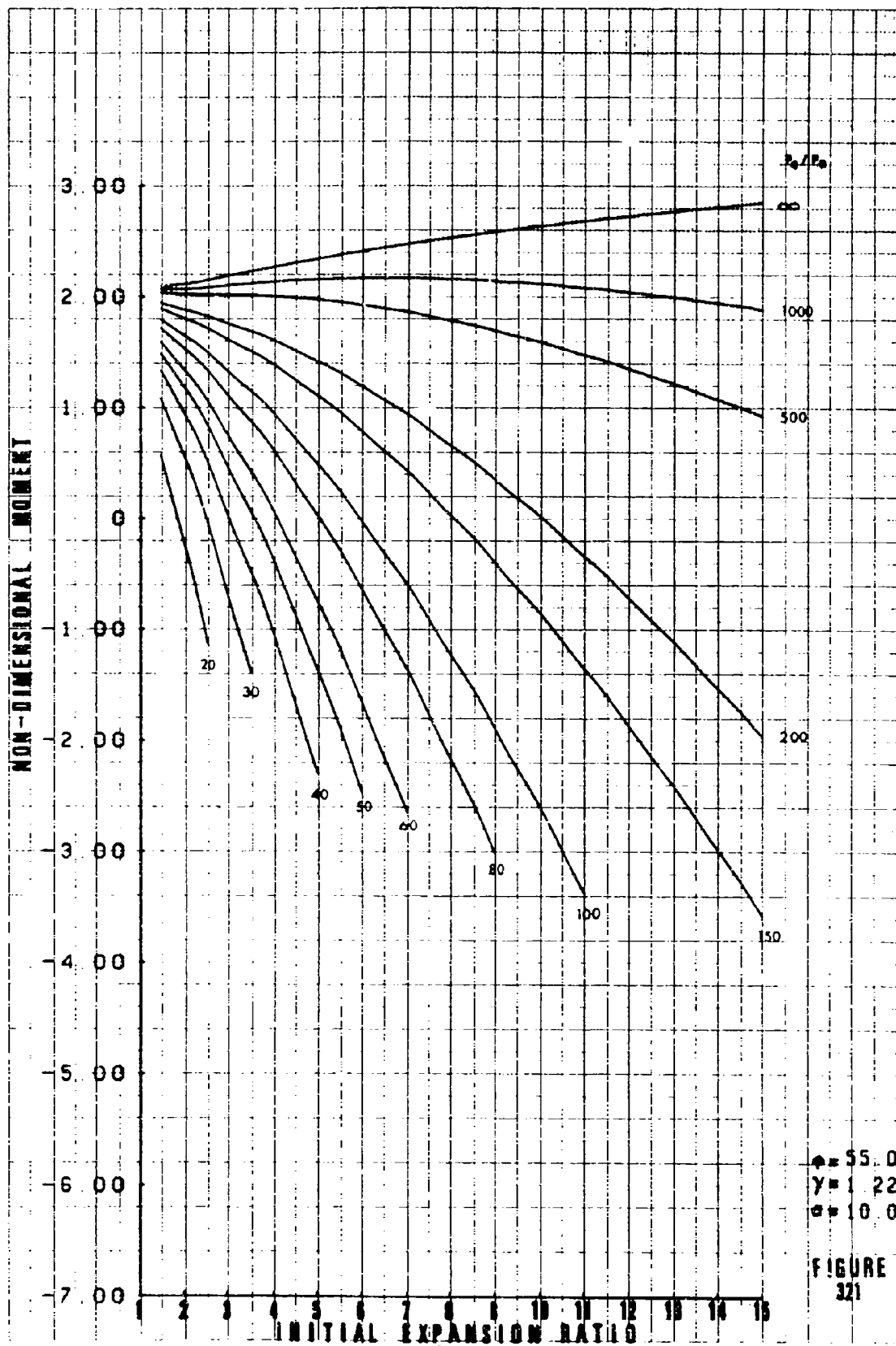


FIGURE 321

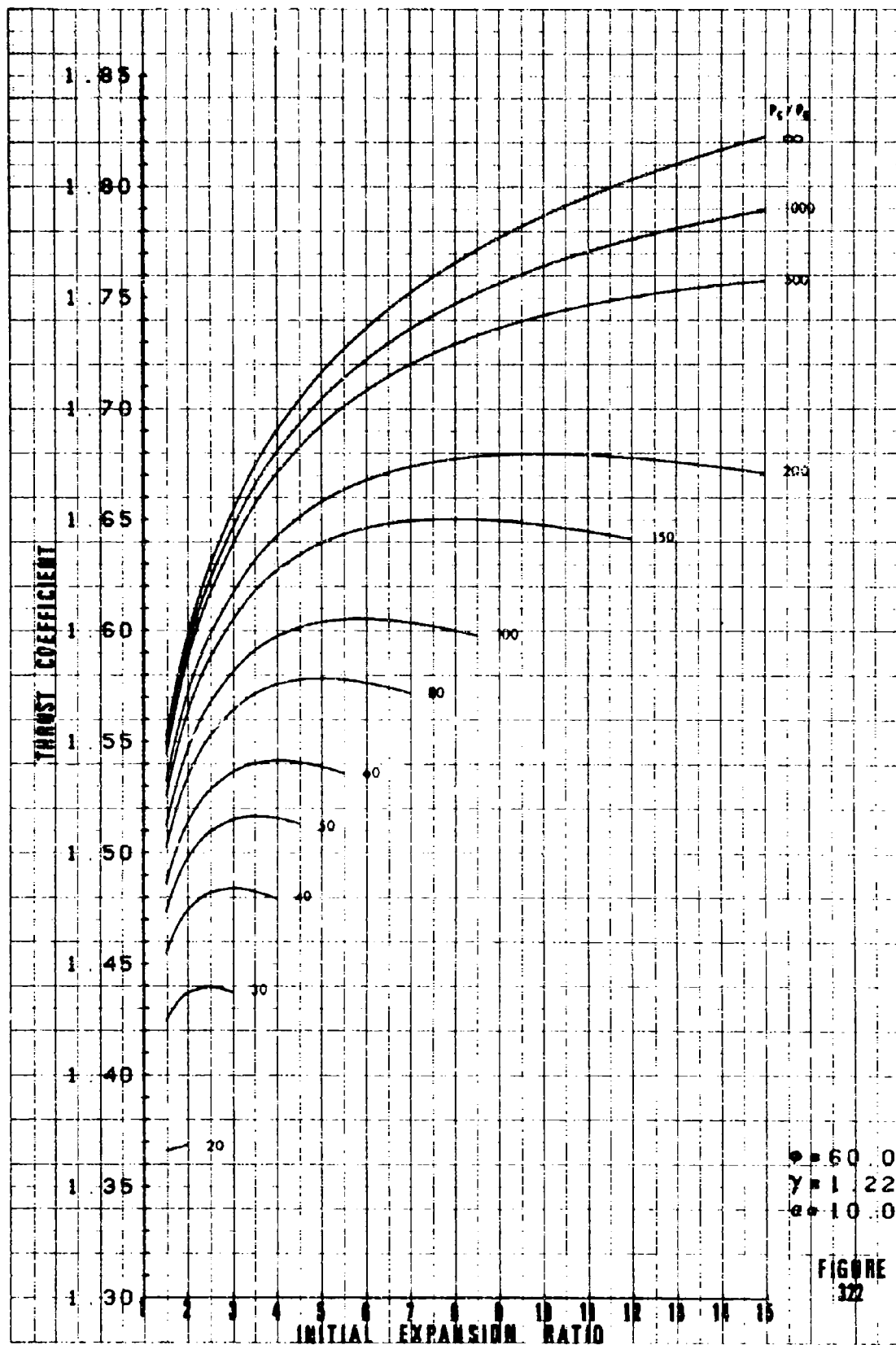


FIGURE 322

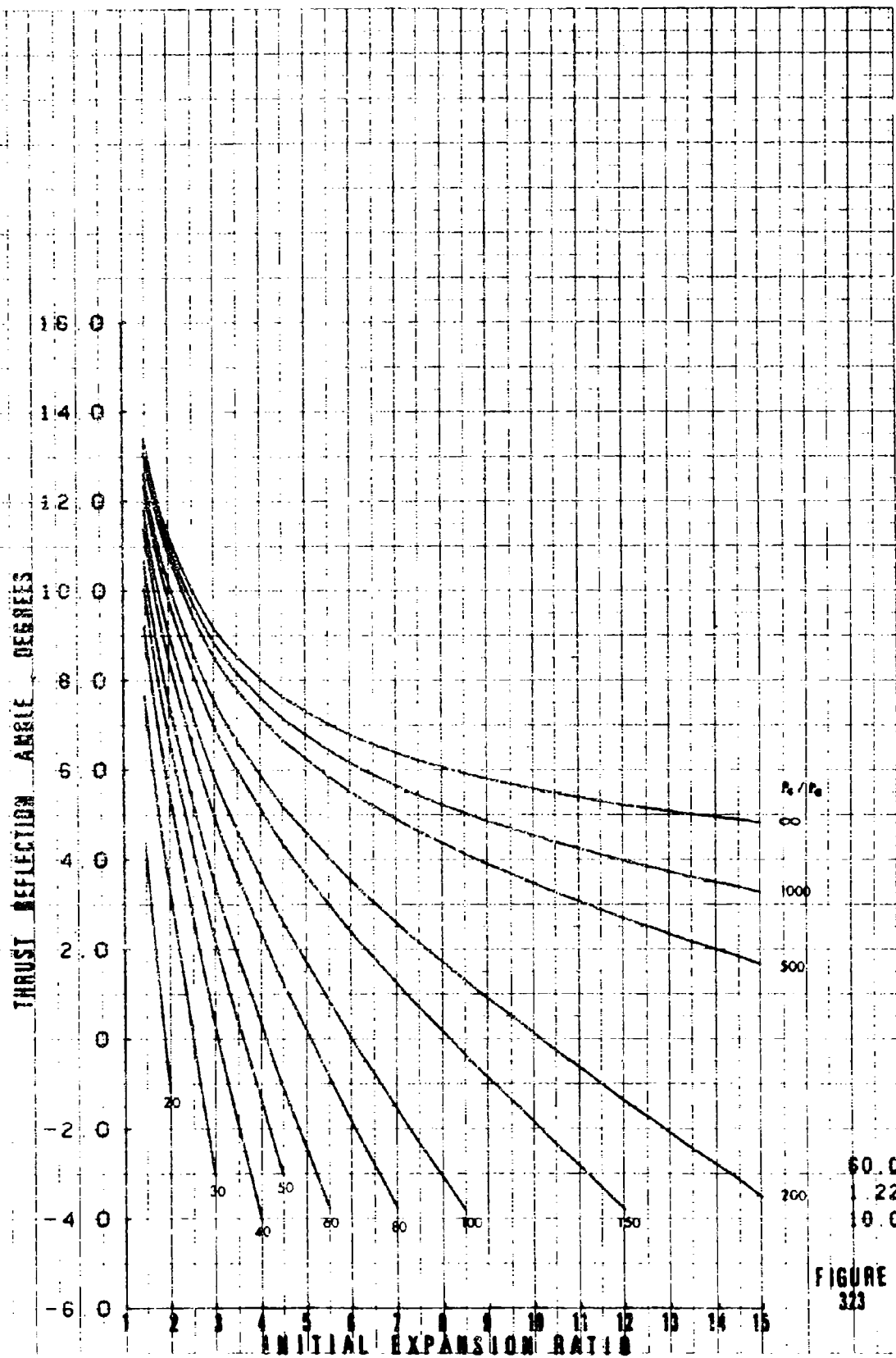
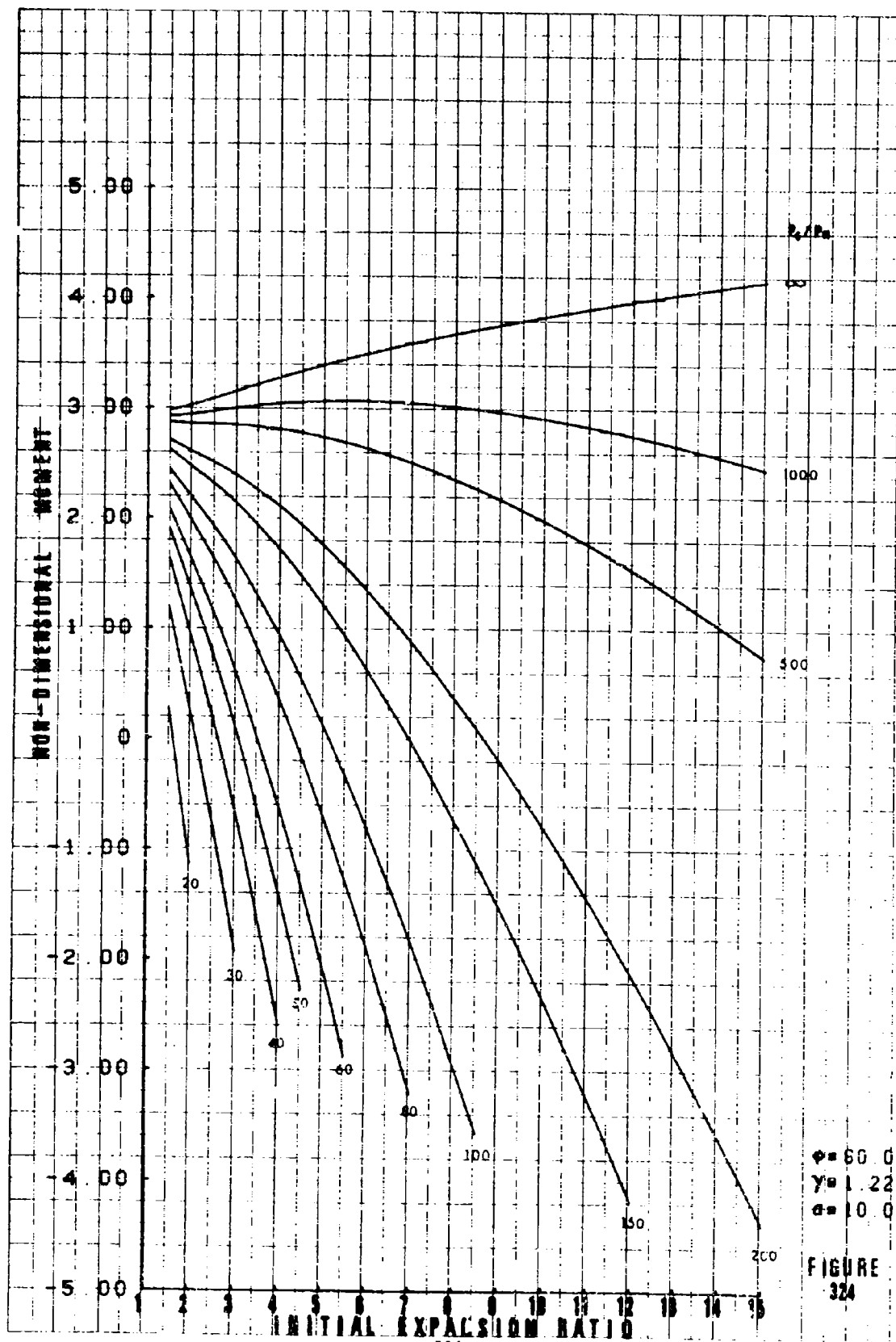
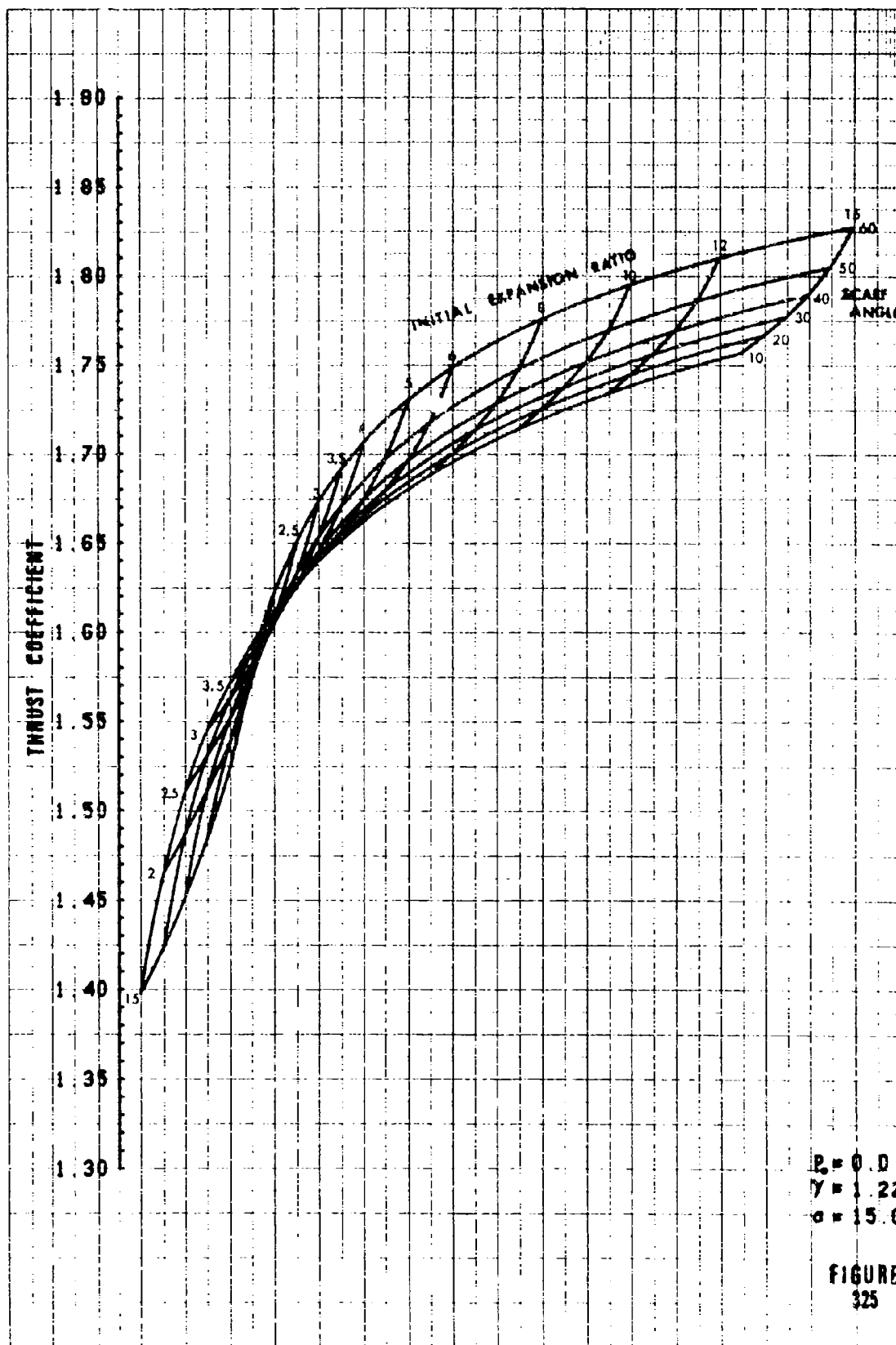
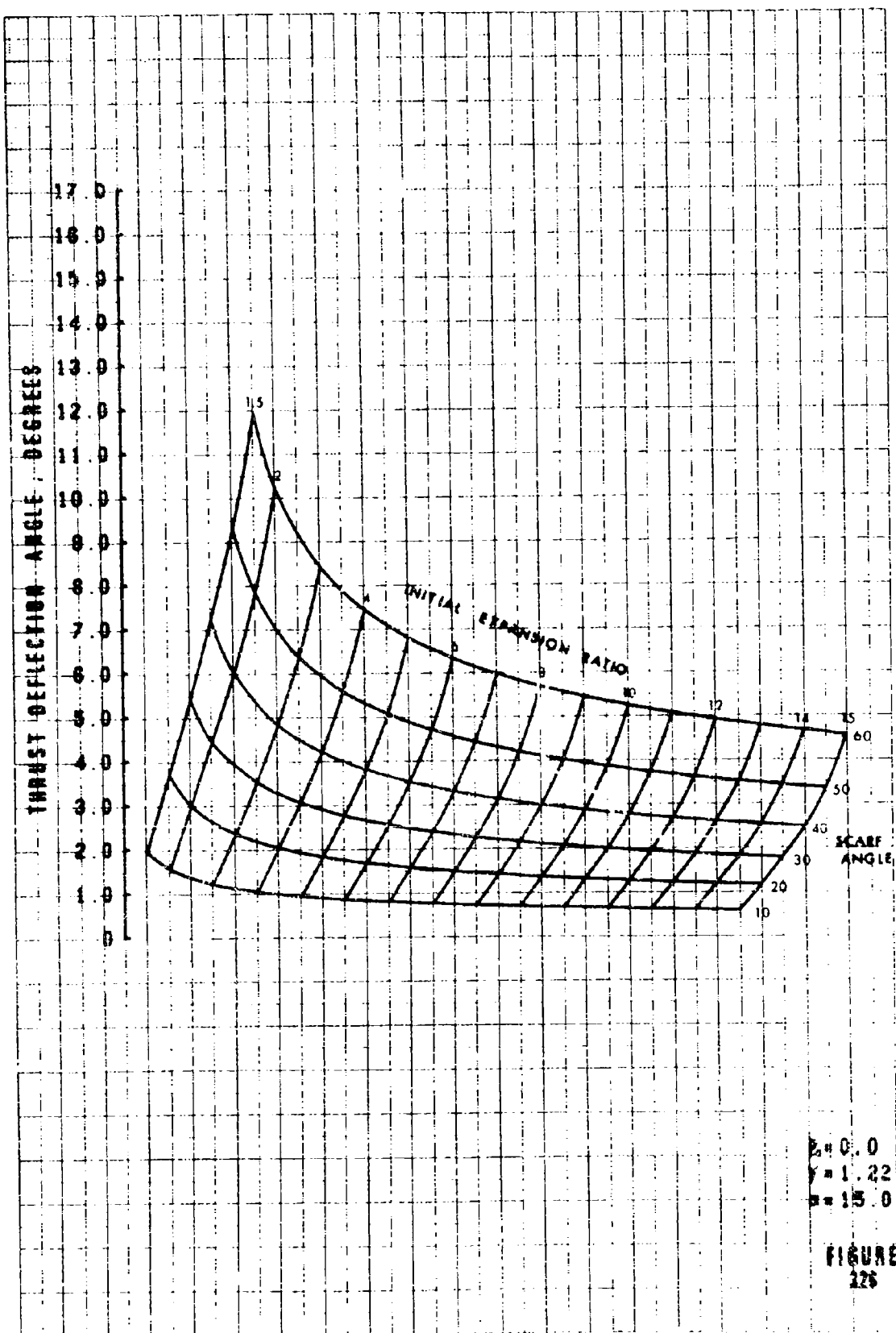


FIGURE 323

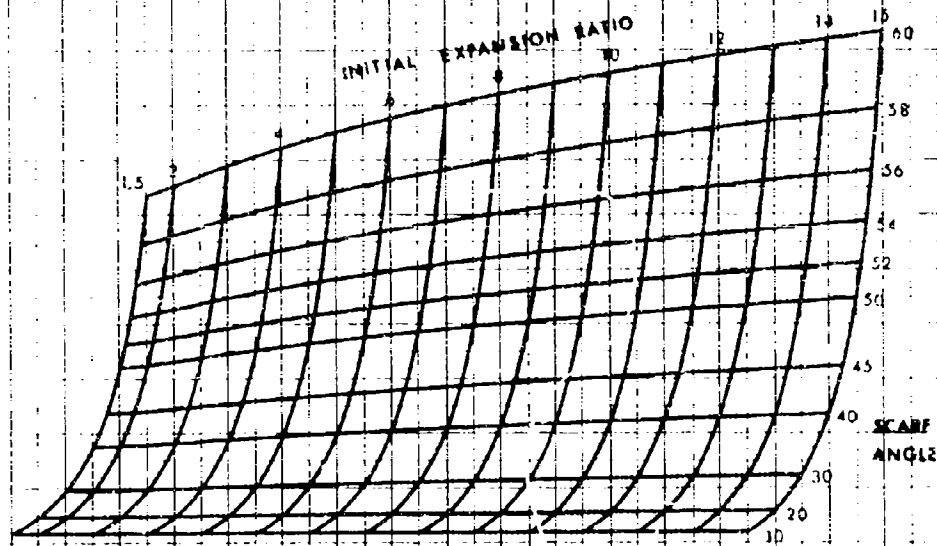






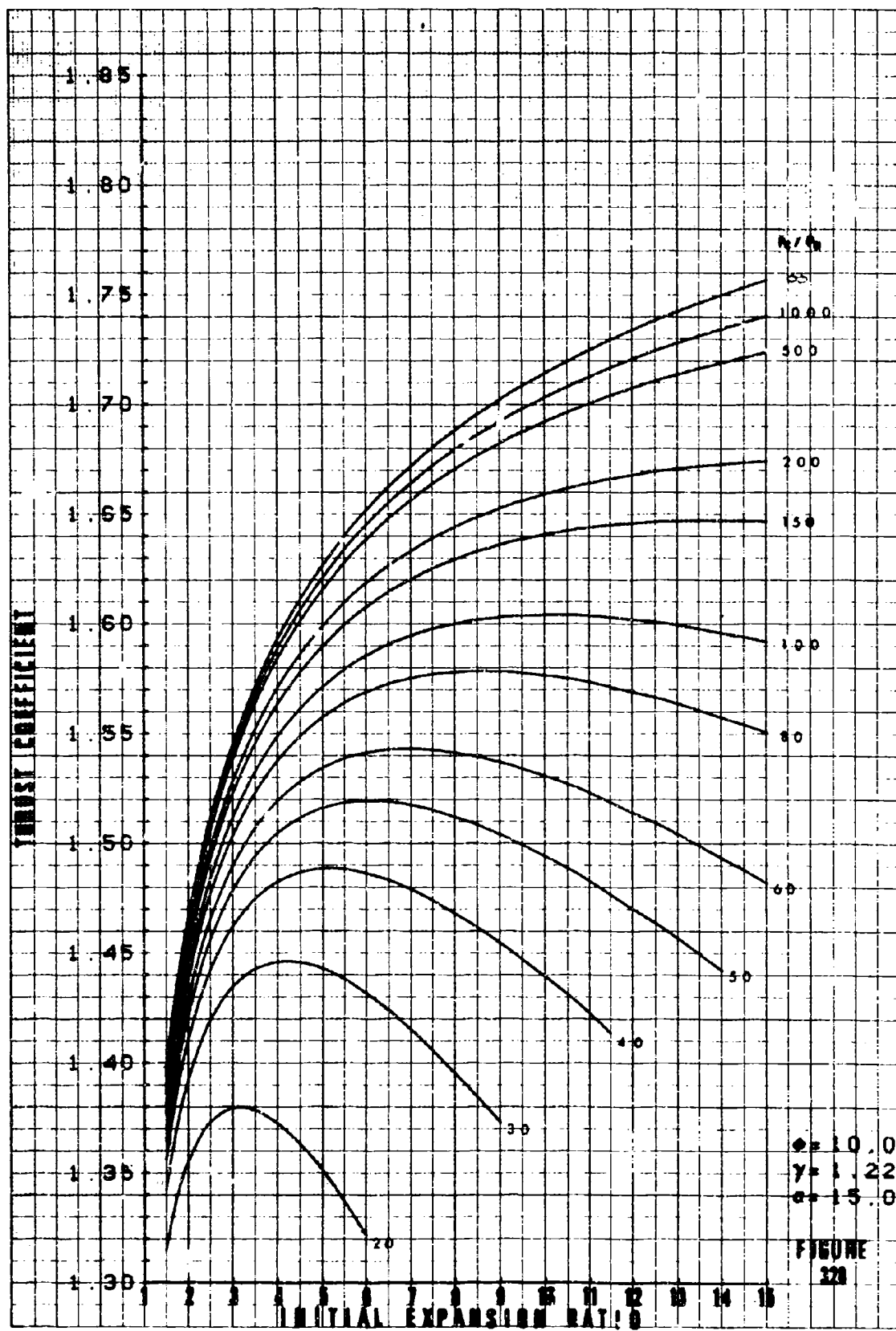
NON-DIMENSIONAL MOMENT

10.0
9.0
8.0
7.0
6.0
5.0
4.0
3.0
2.0
1.0
0



$P_0 = 0.0$
 $\gamma = 1.22$
 $\alpha = 5.0$

FIGURE
327



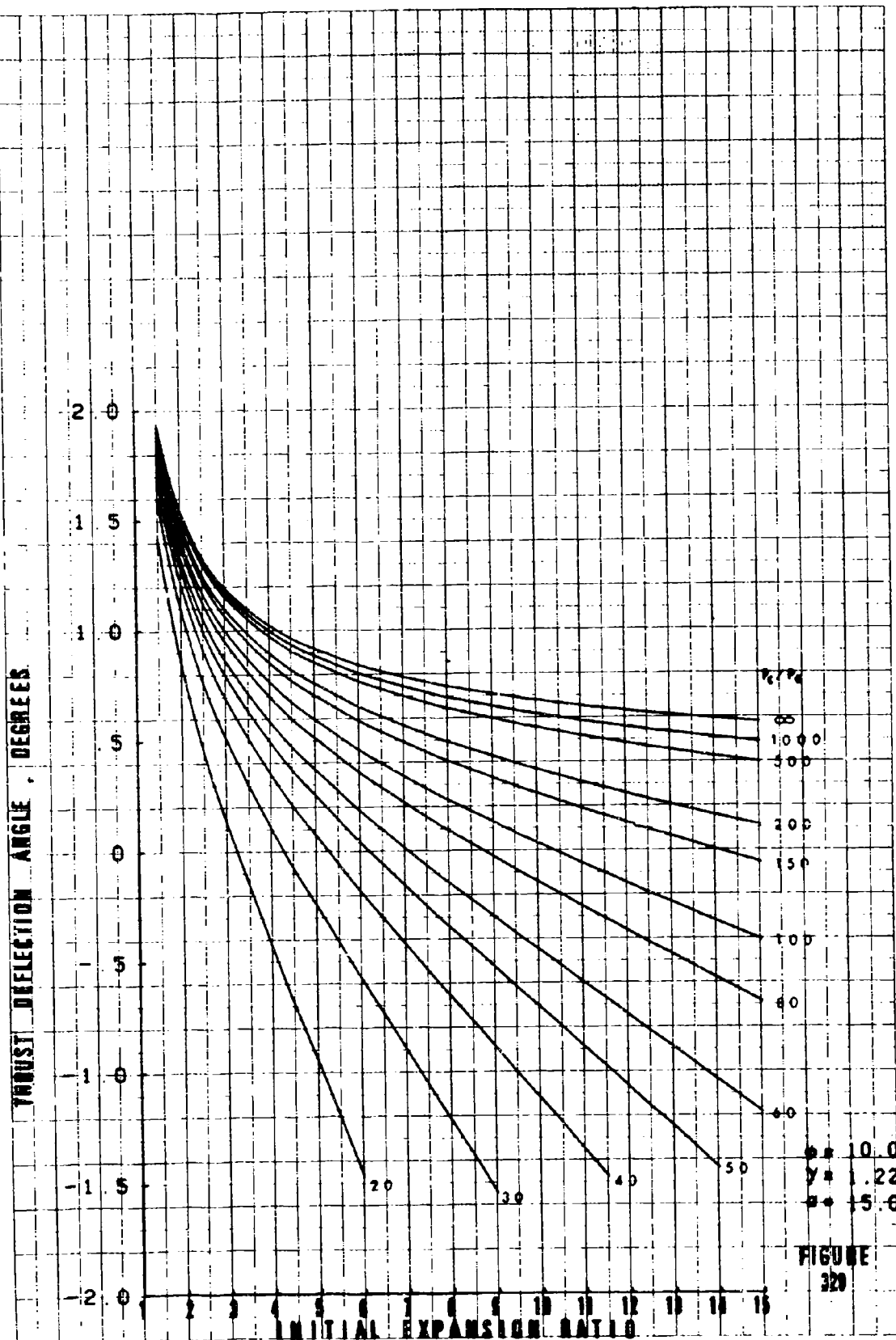
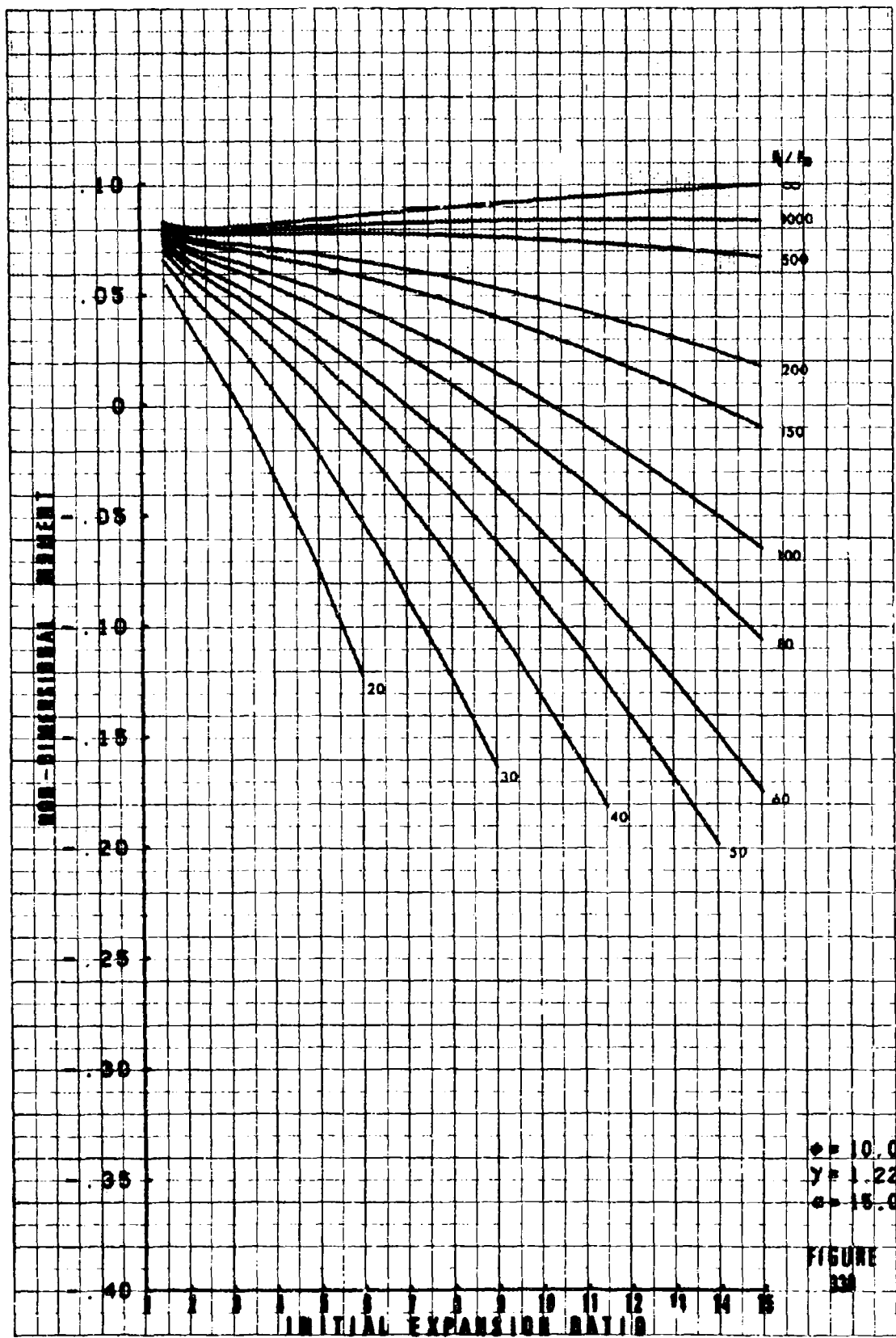


FIGURE 320



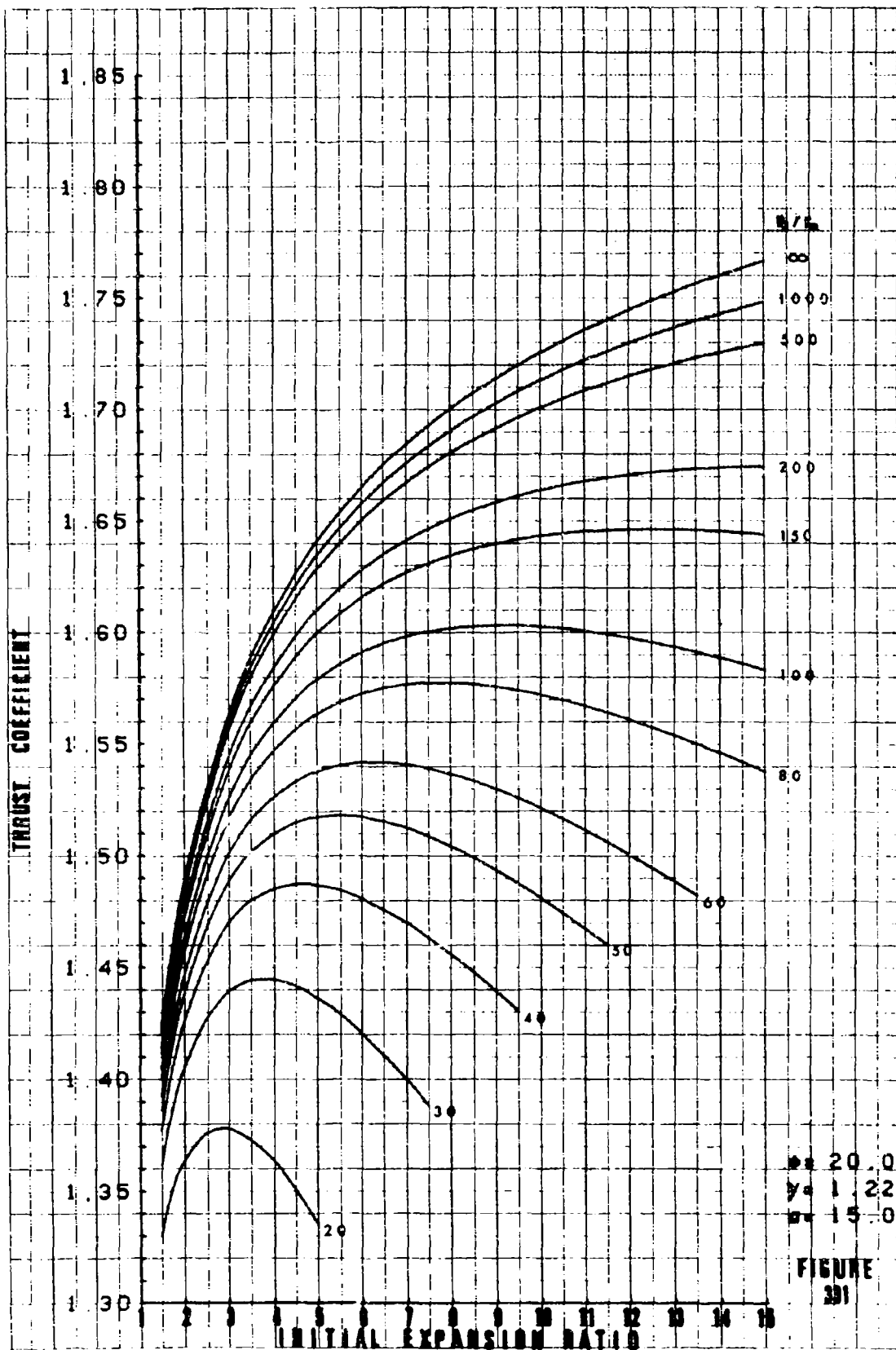


FIGURE
331

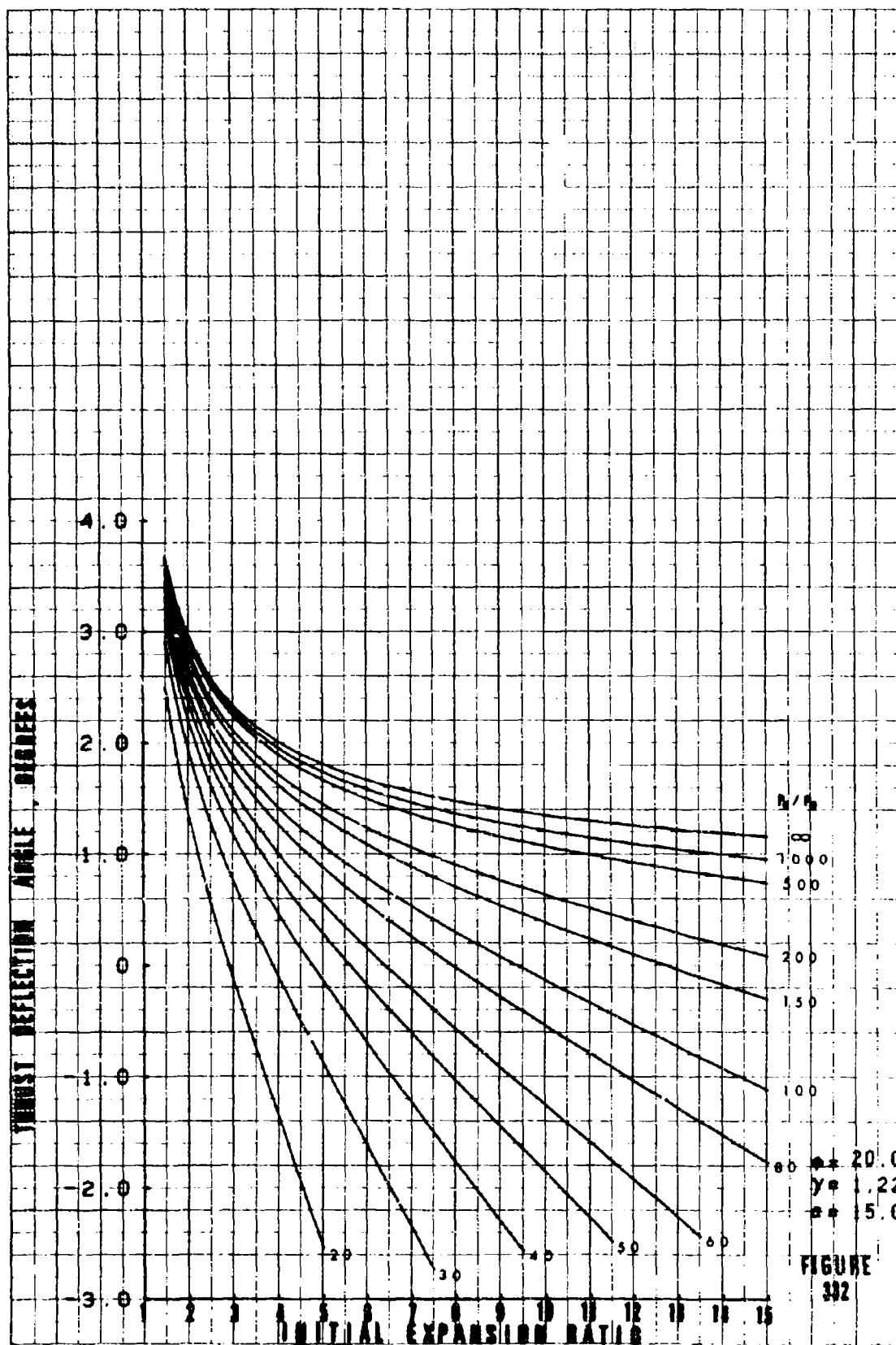
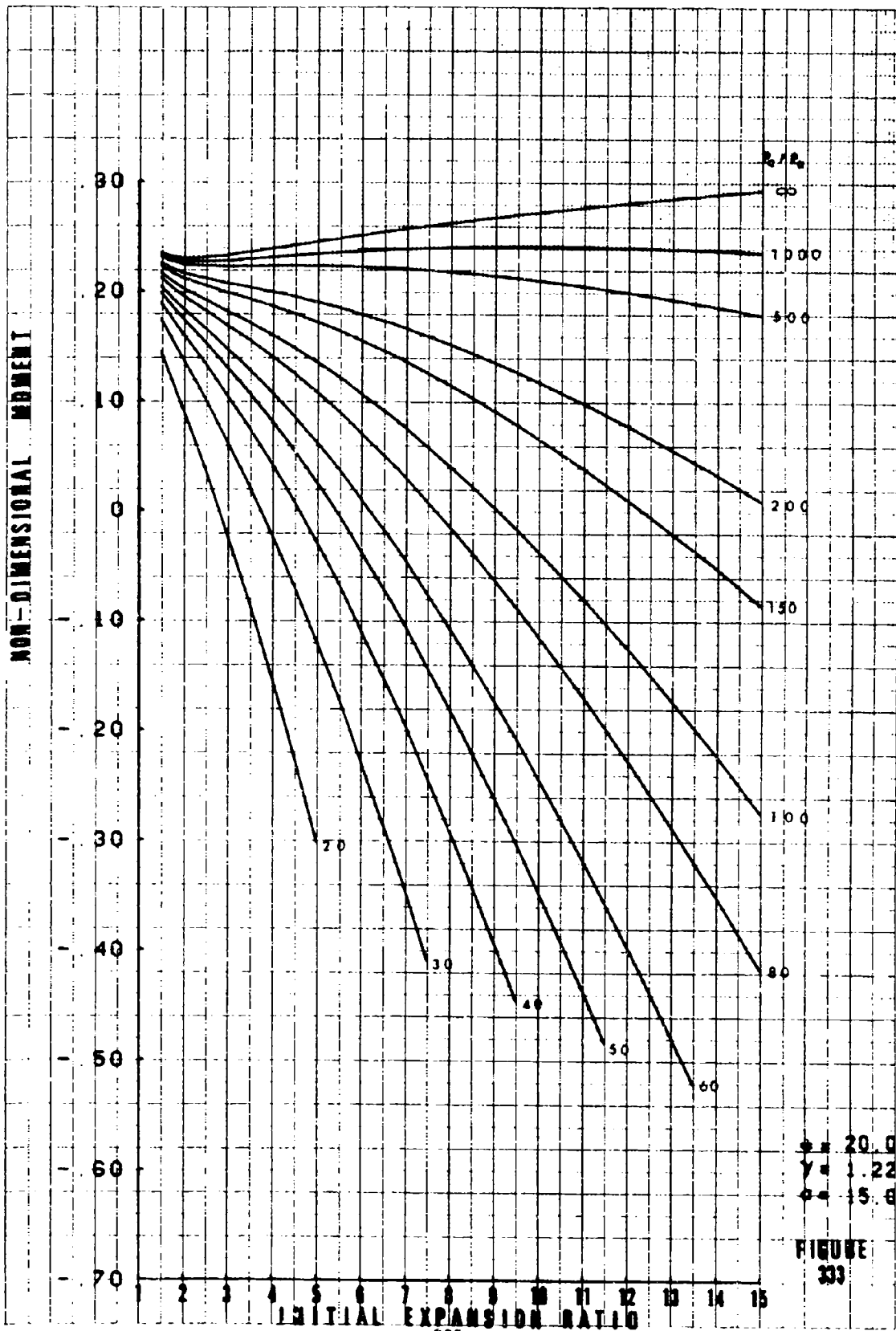
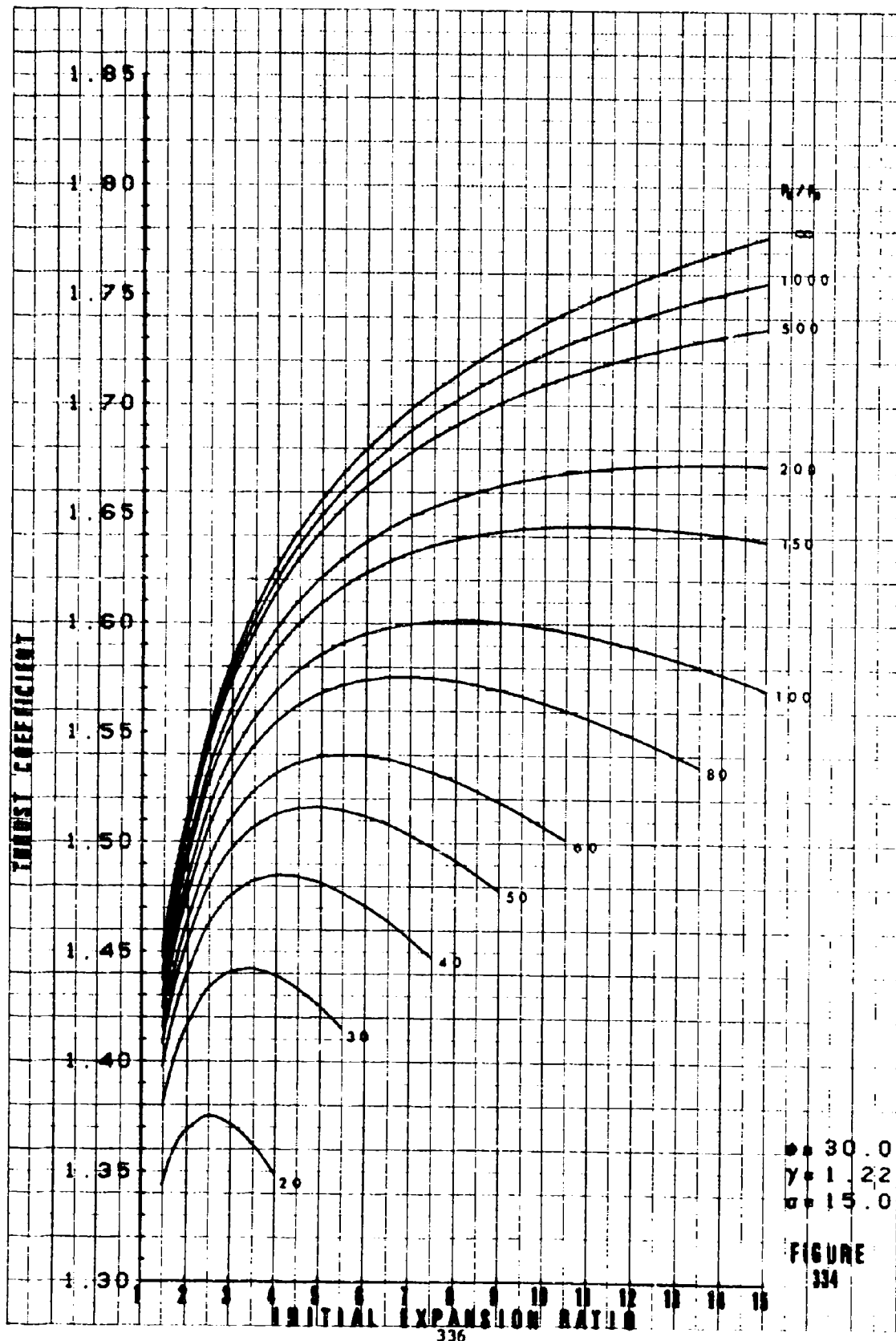
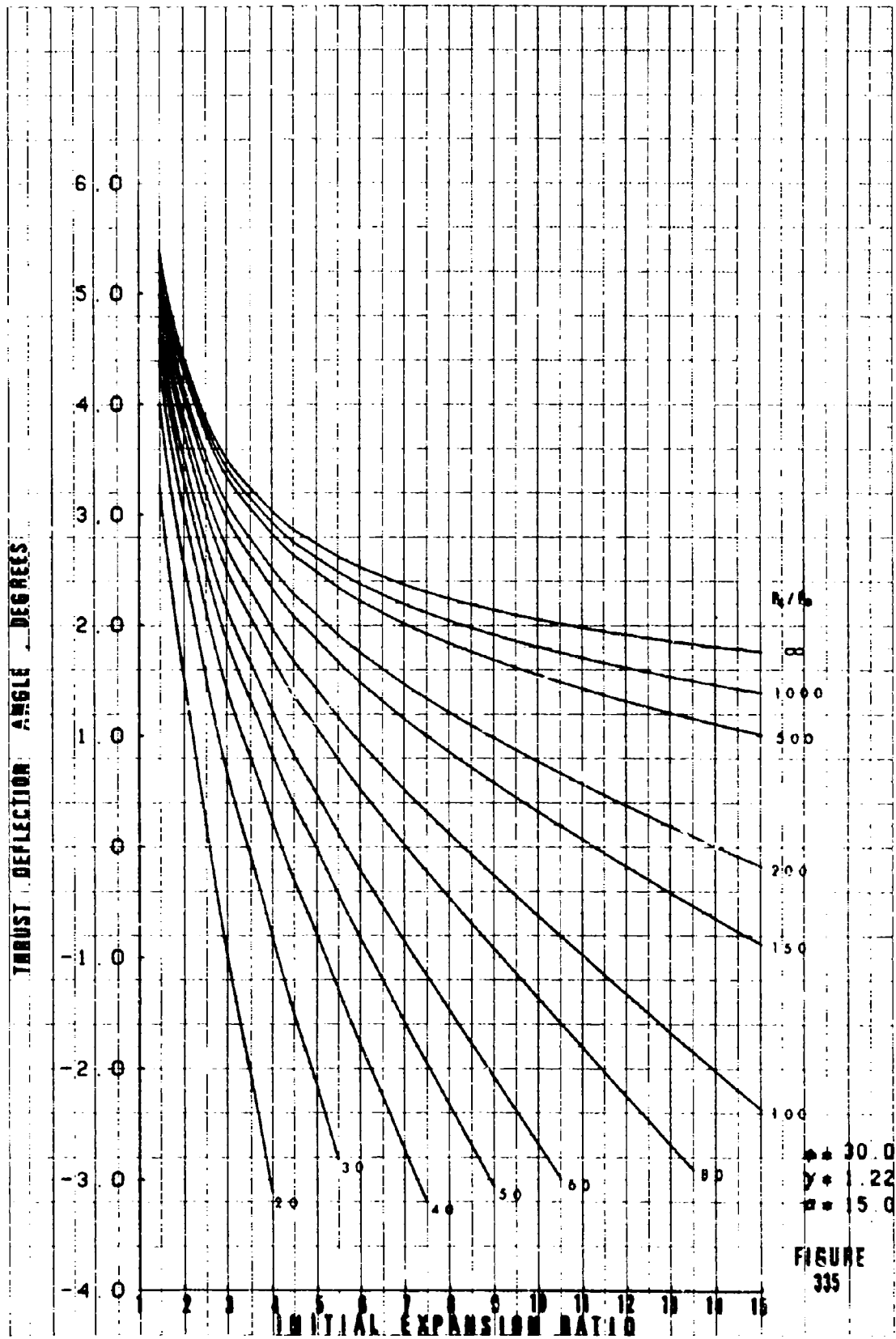
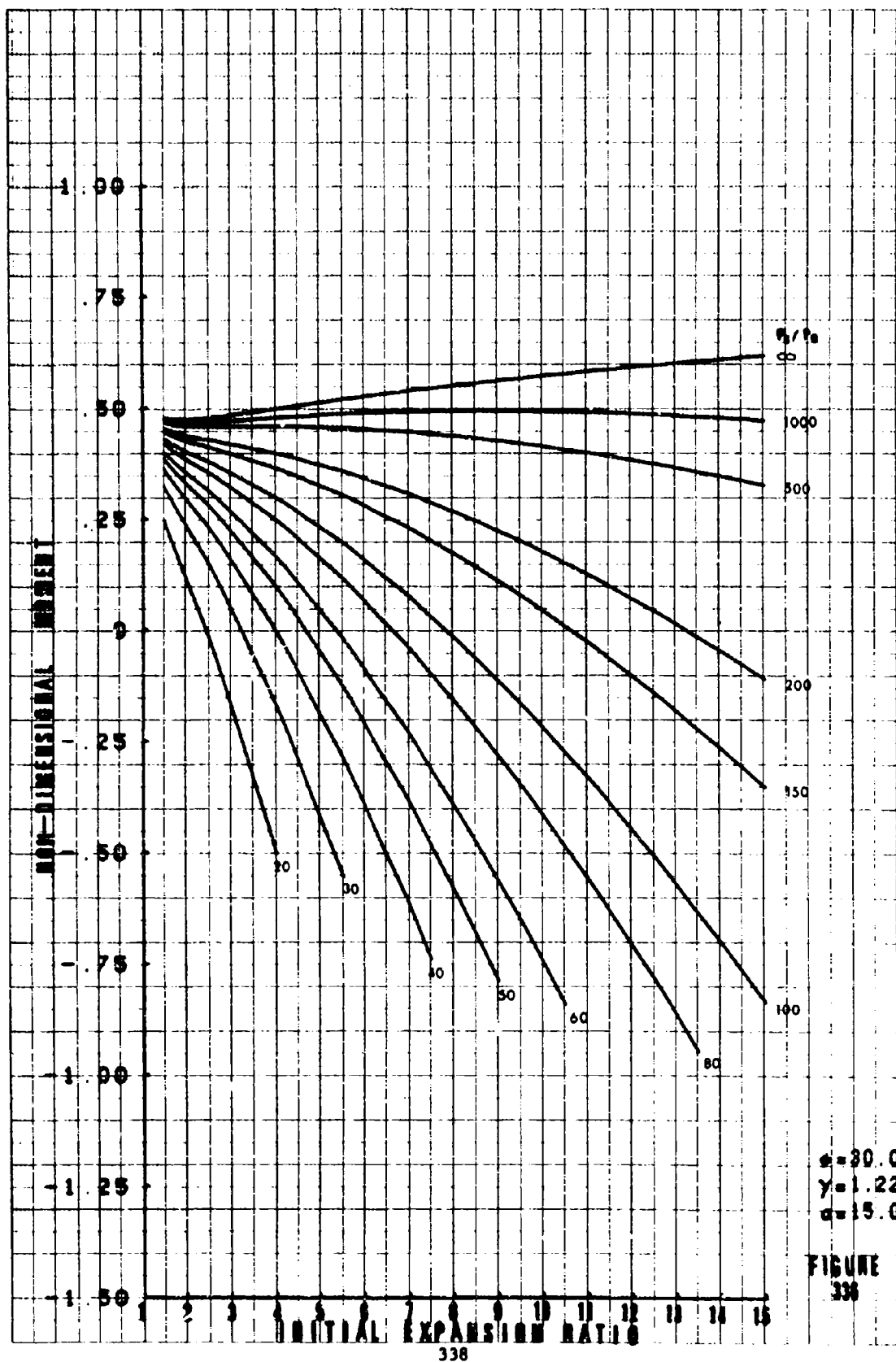


FIGURE 302

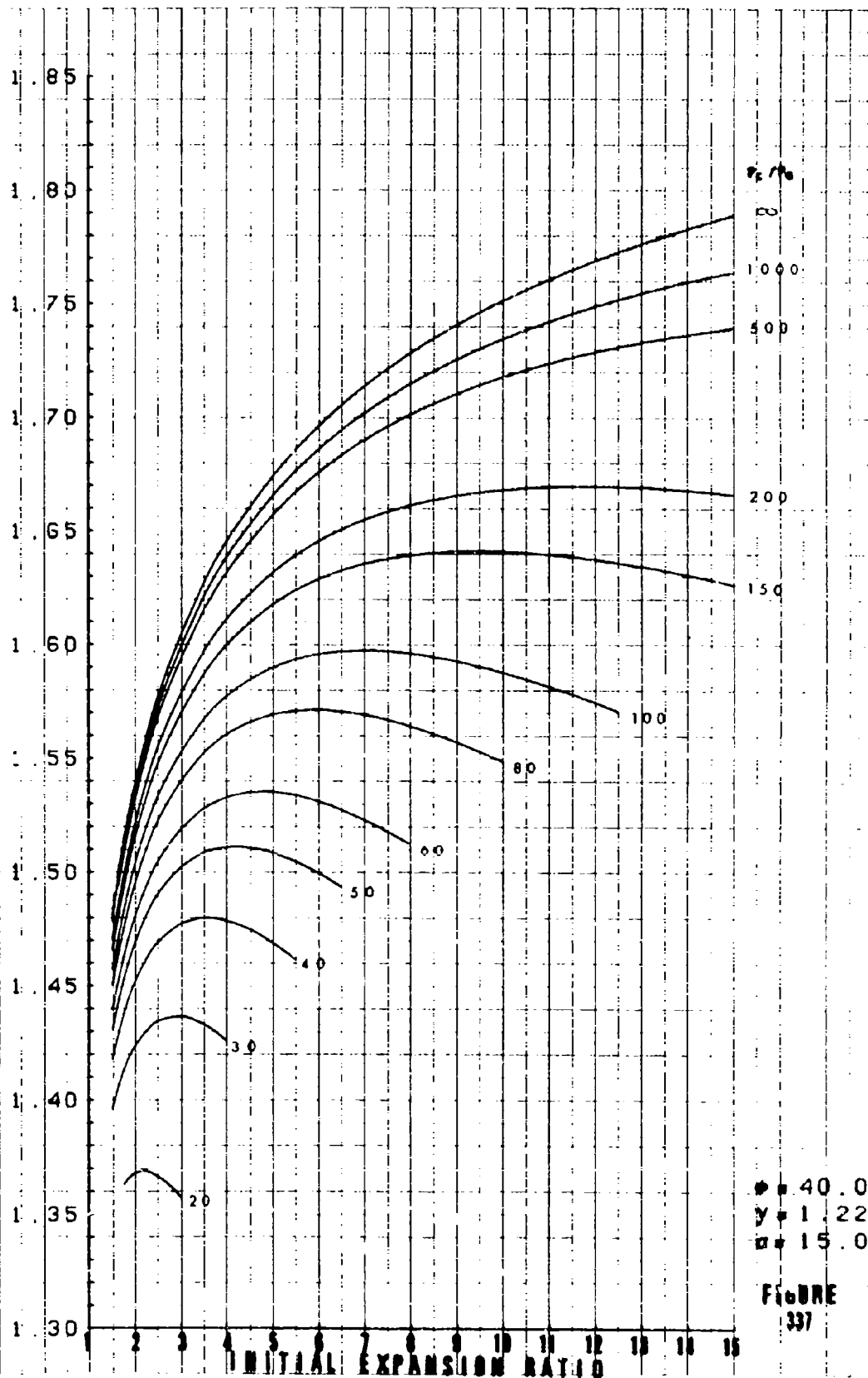






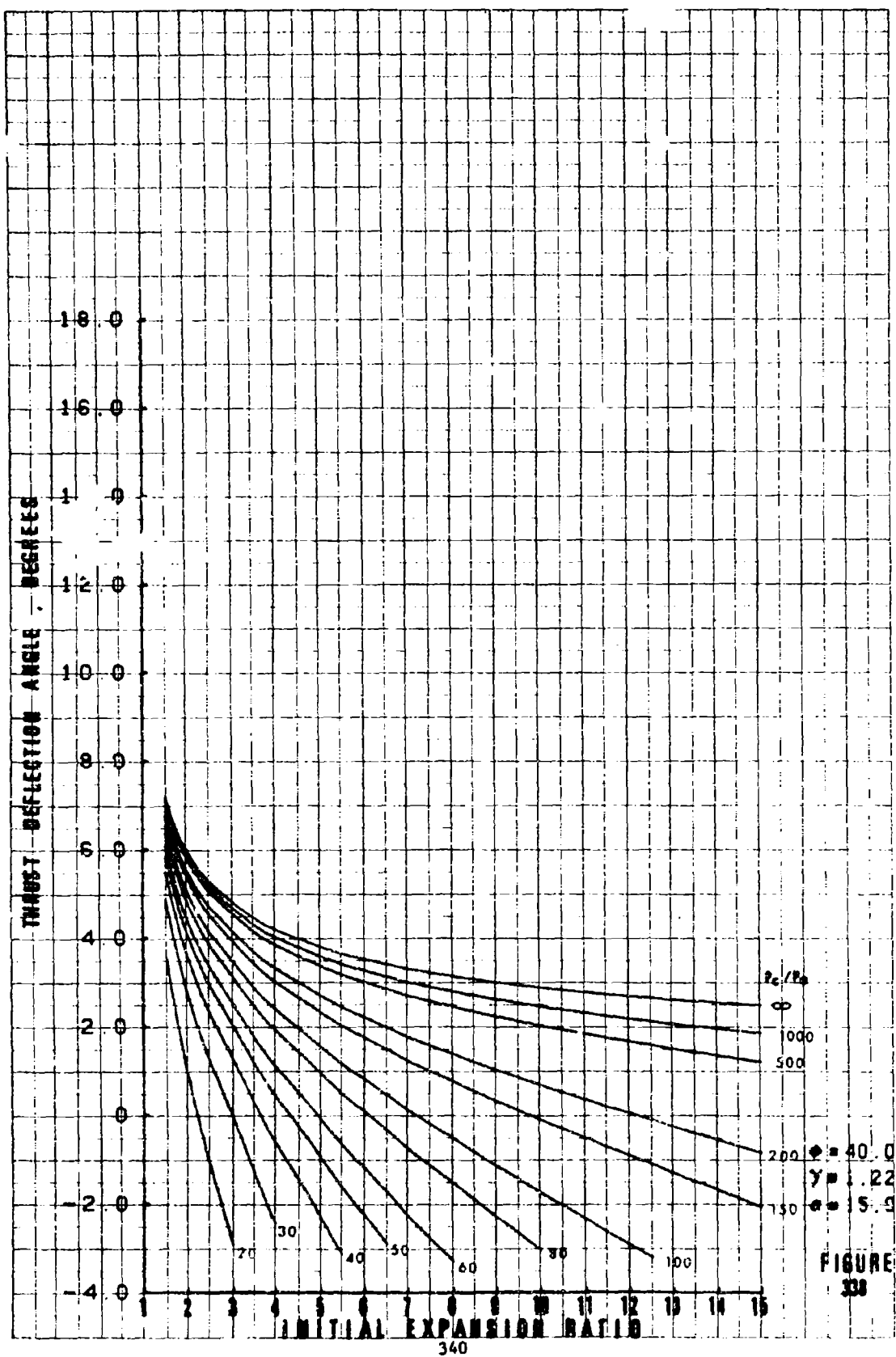


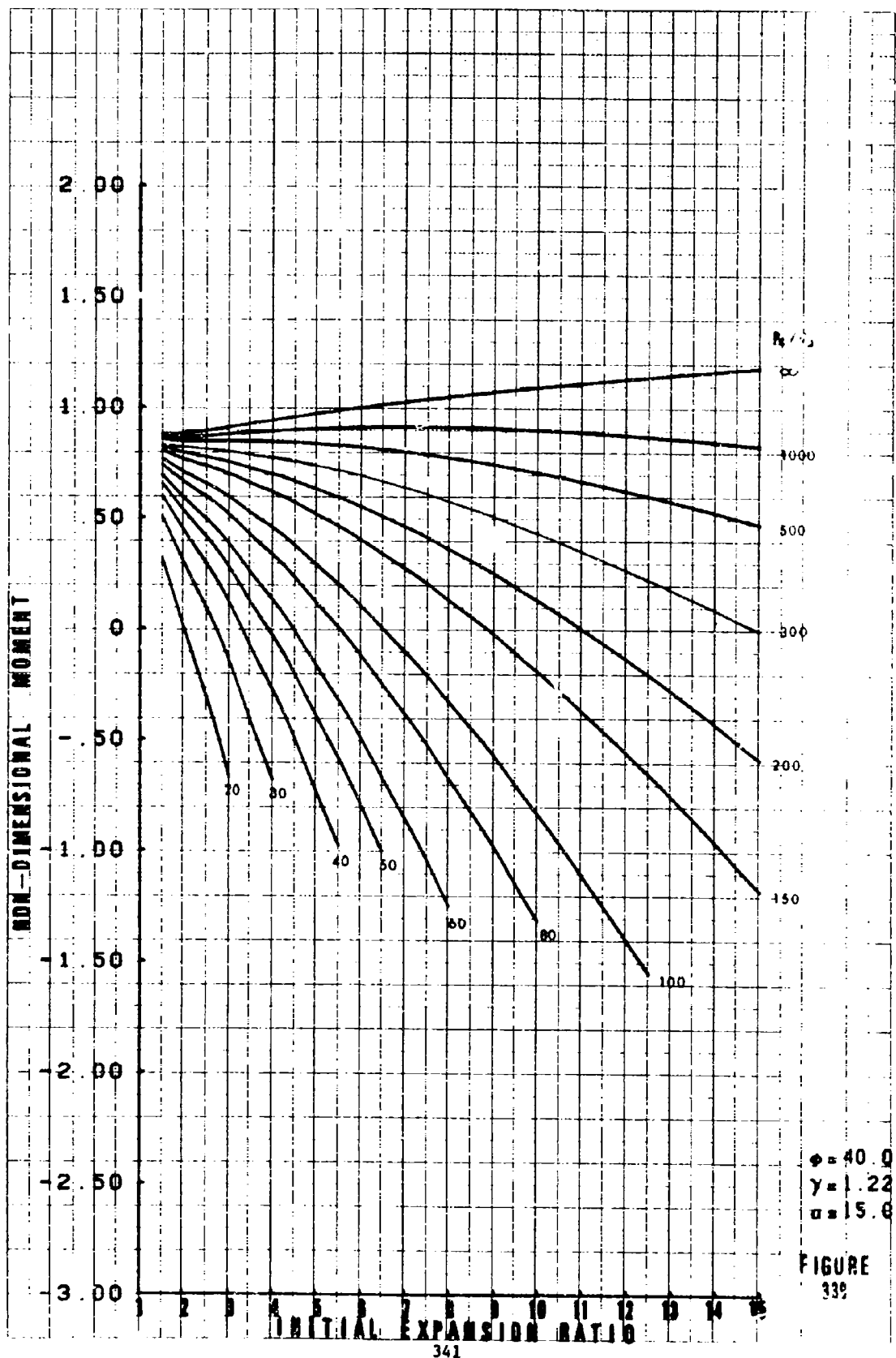
THRUST COEFFICIENT

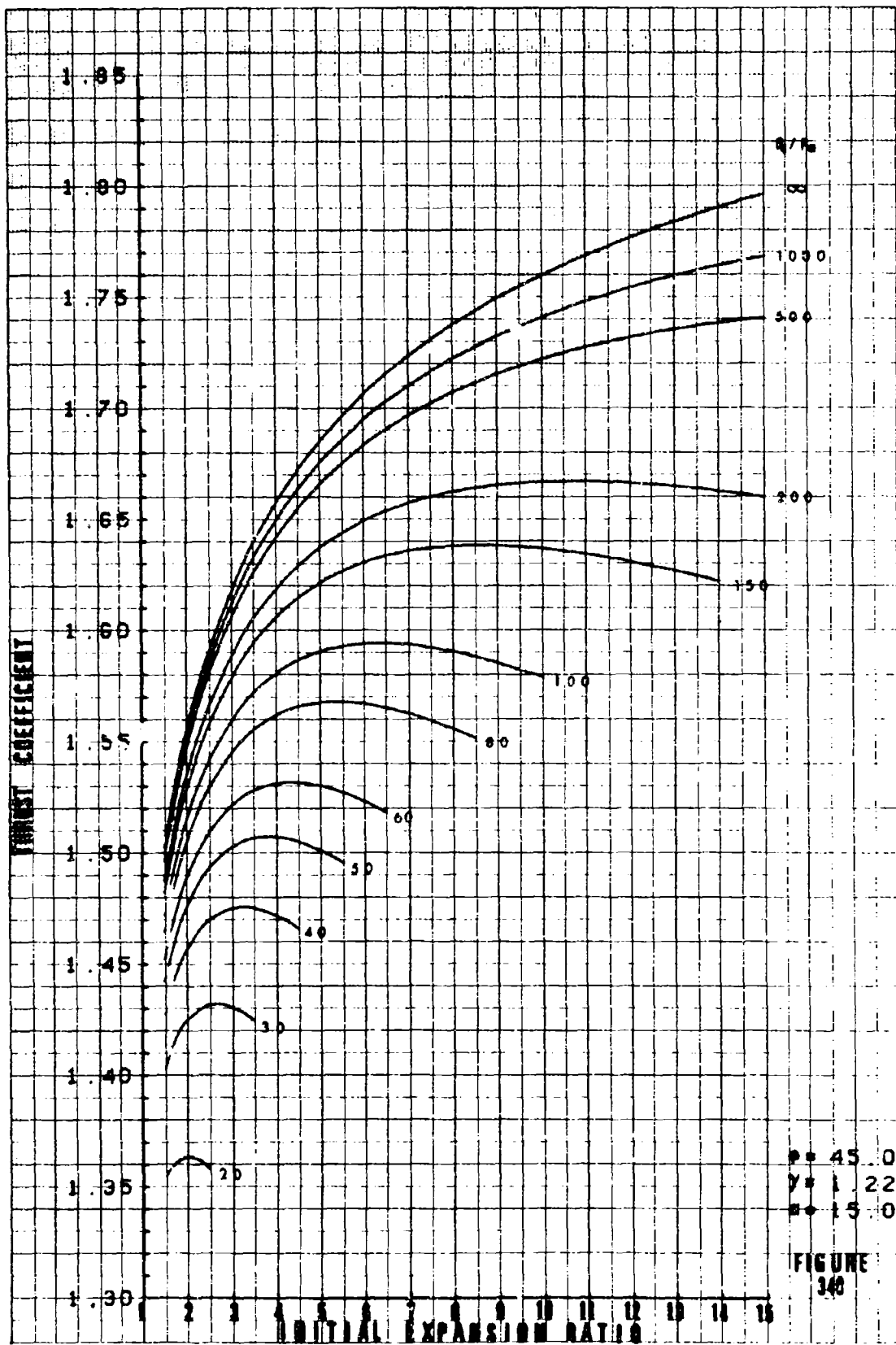


$\gamma = 40.0$
 $\gamma = 1.22$
 $\gamma = 15.0$

FIGURE 337







P = 45.0
 Y = 1.22
 Z = 15.0

FIGURE 340

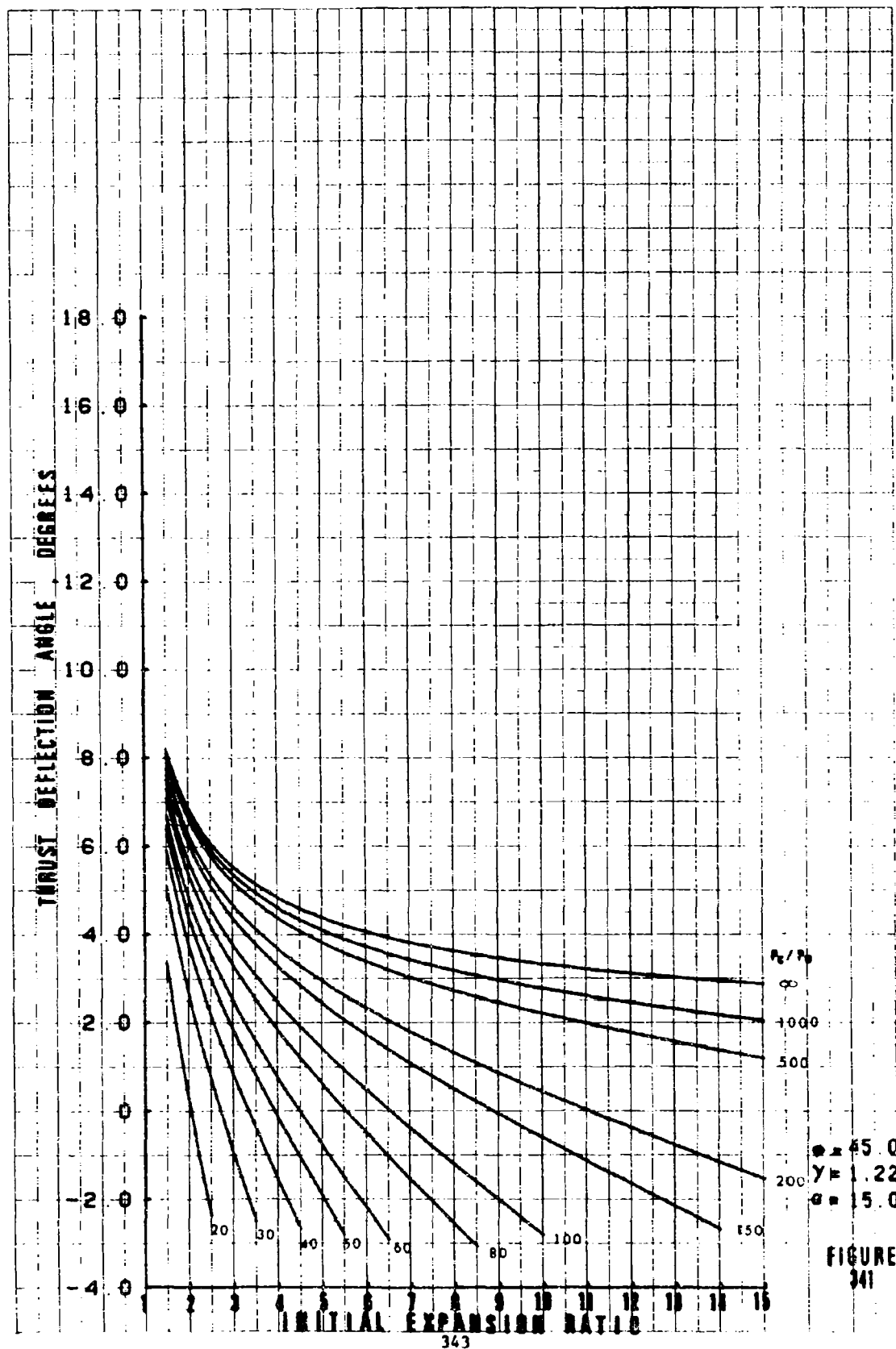
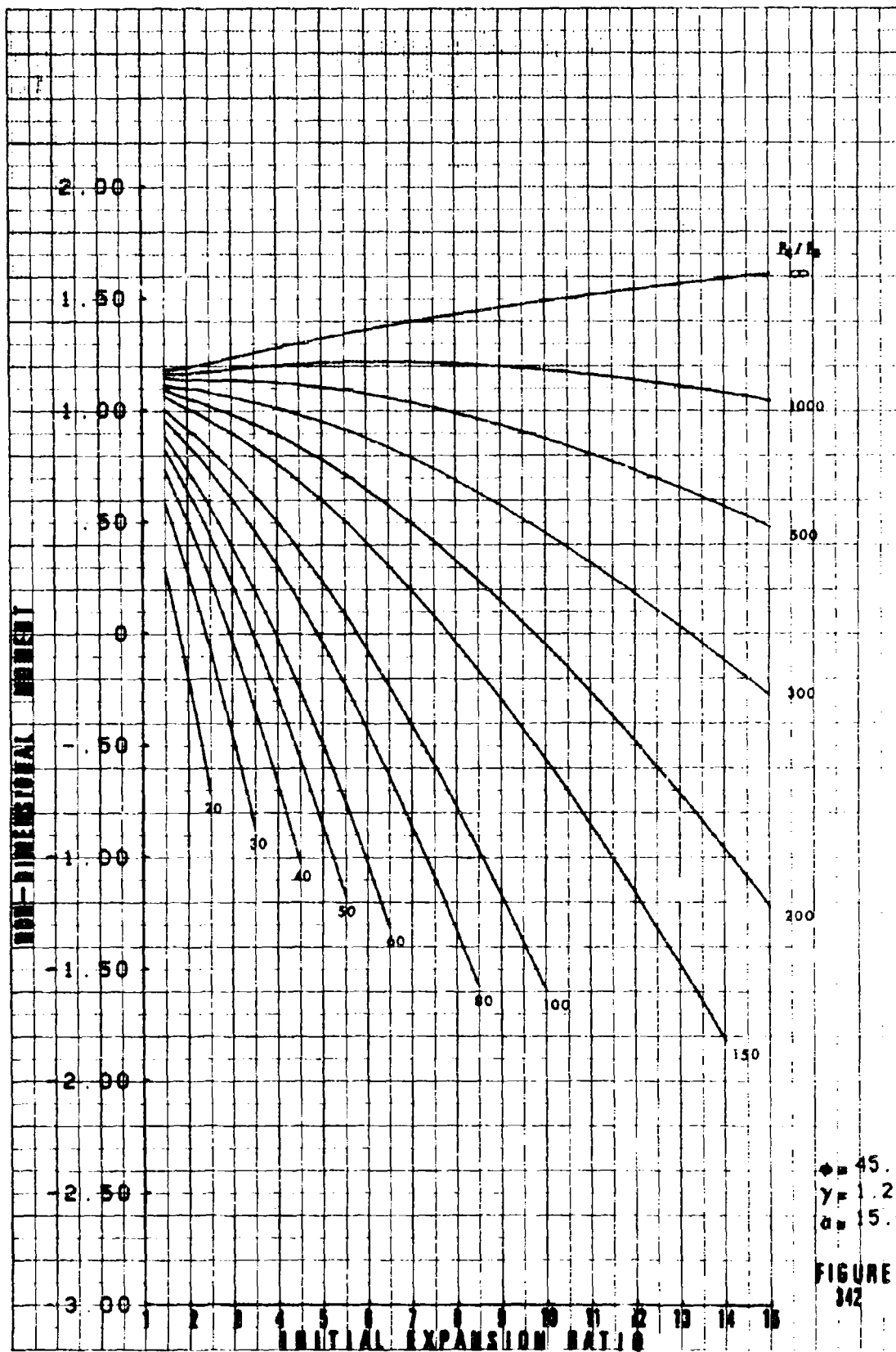
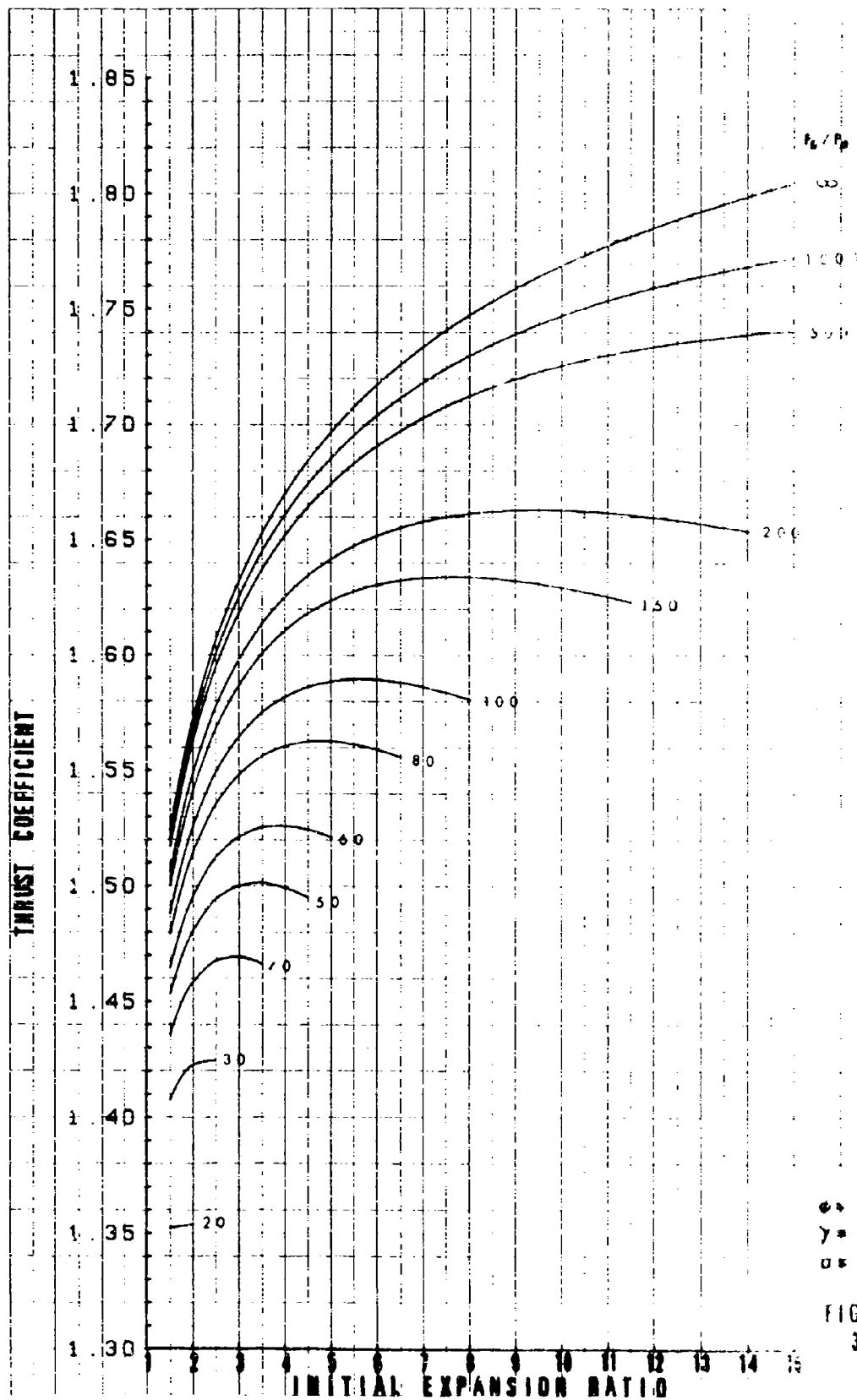


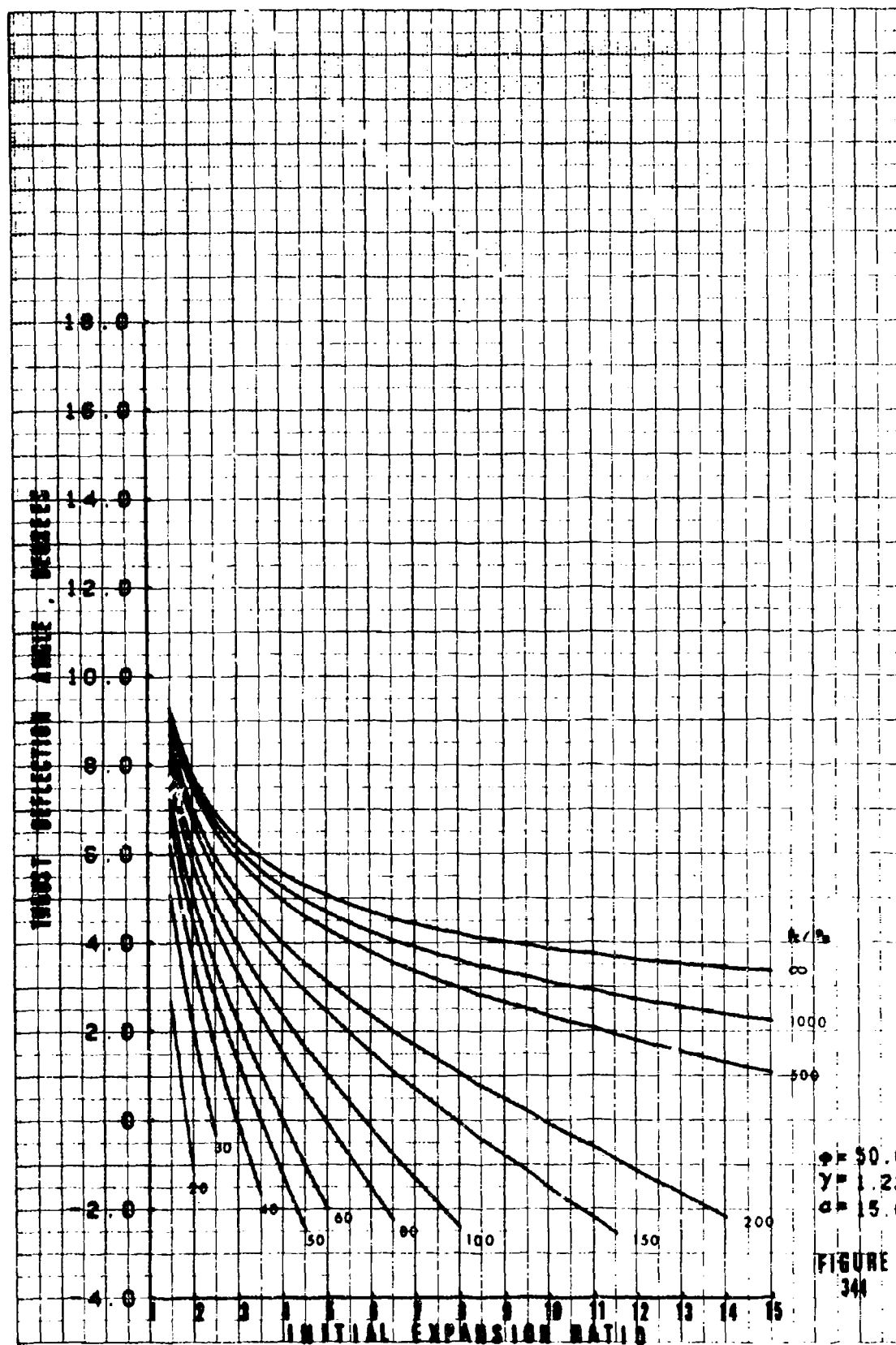
FIGURE 341

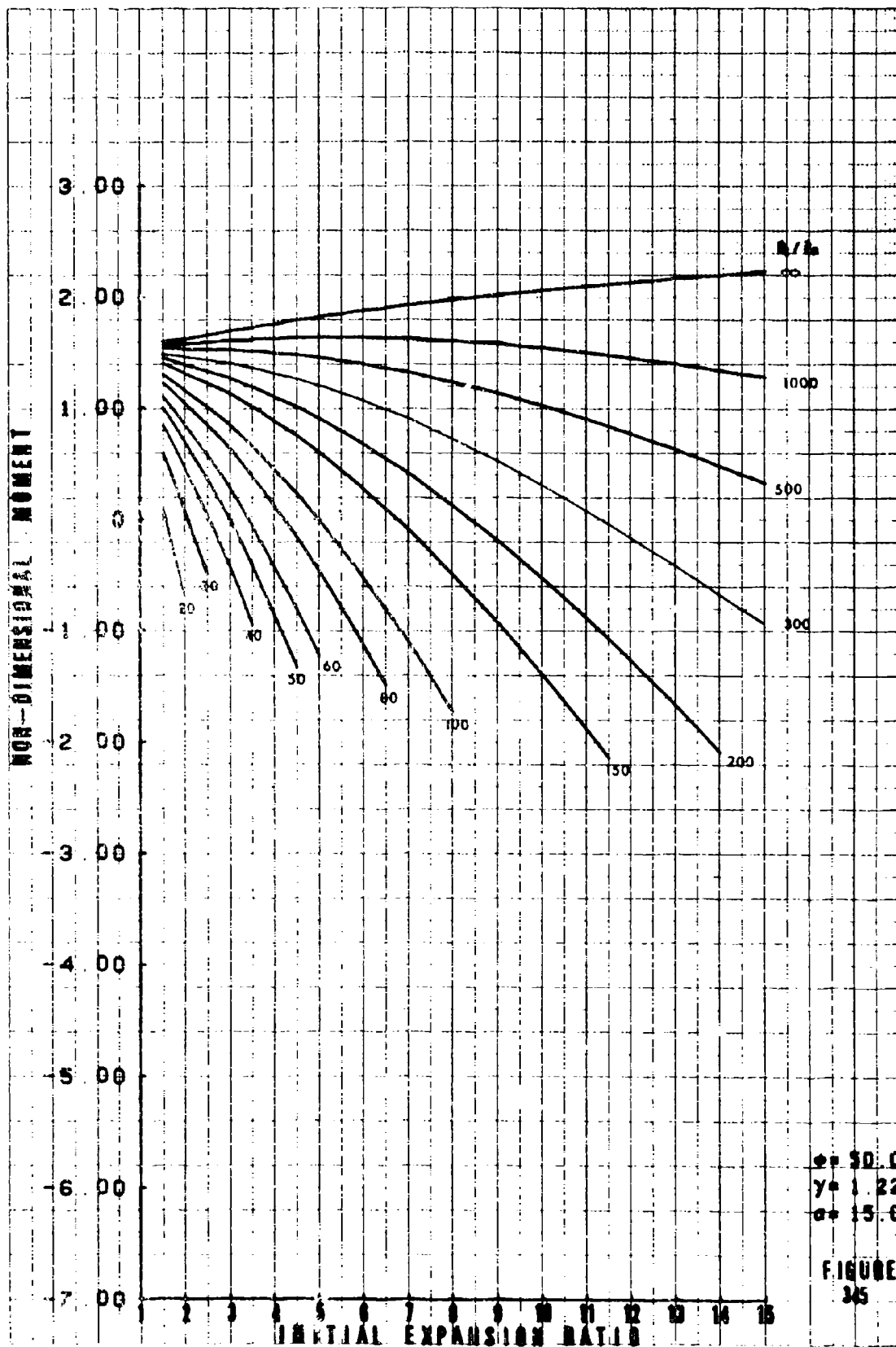


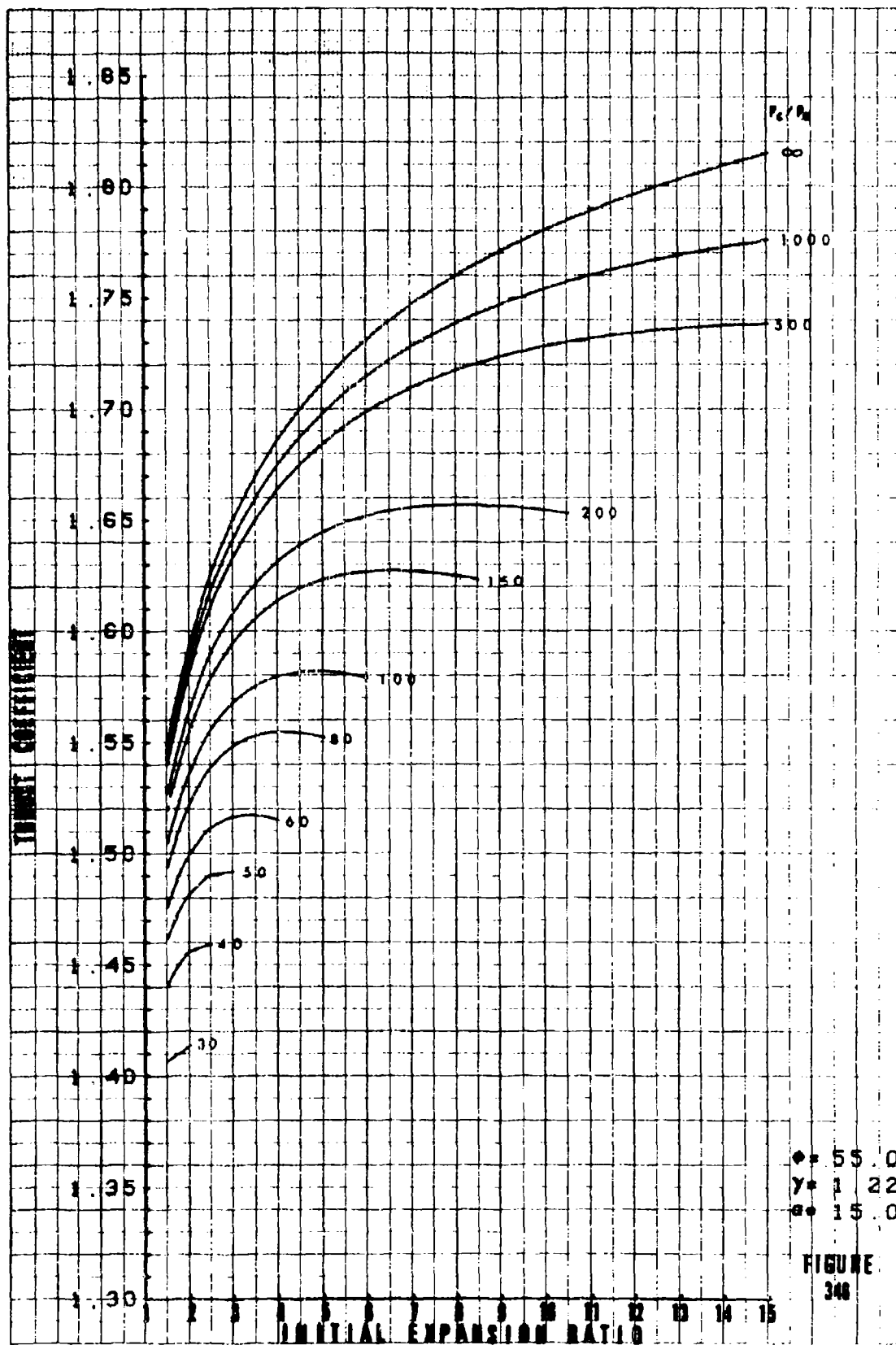


$\phi = 50.0$
 $\gamma = 1.22$
 $\alpha = 10.0$

FIGURE
343







$\phi = 55.0$
 $\gamma = 1.22$
 $\alpha = 15.0$

FIGURE 348

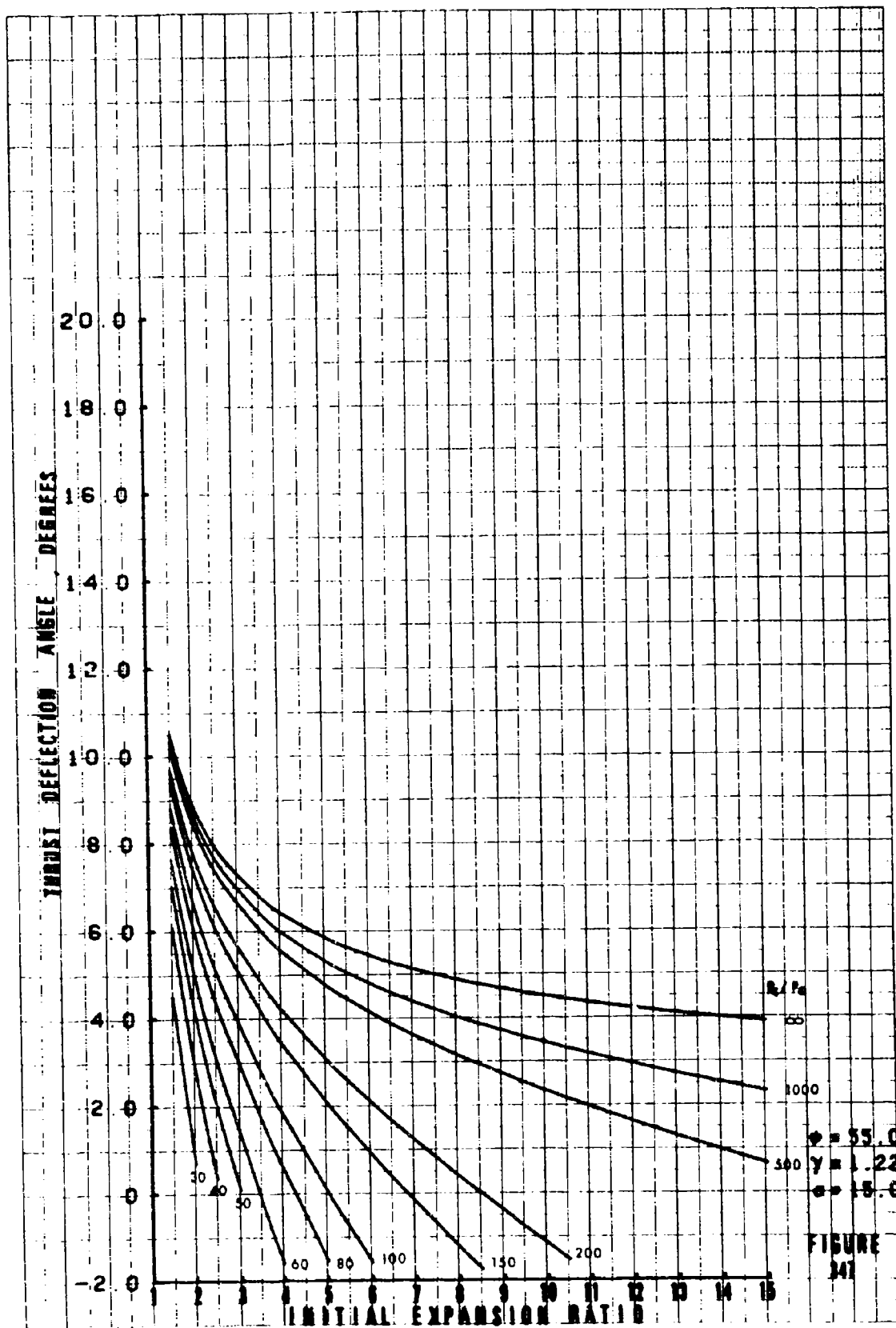
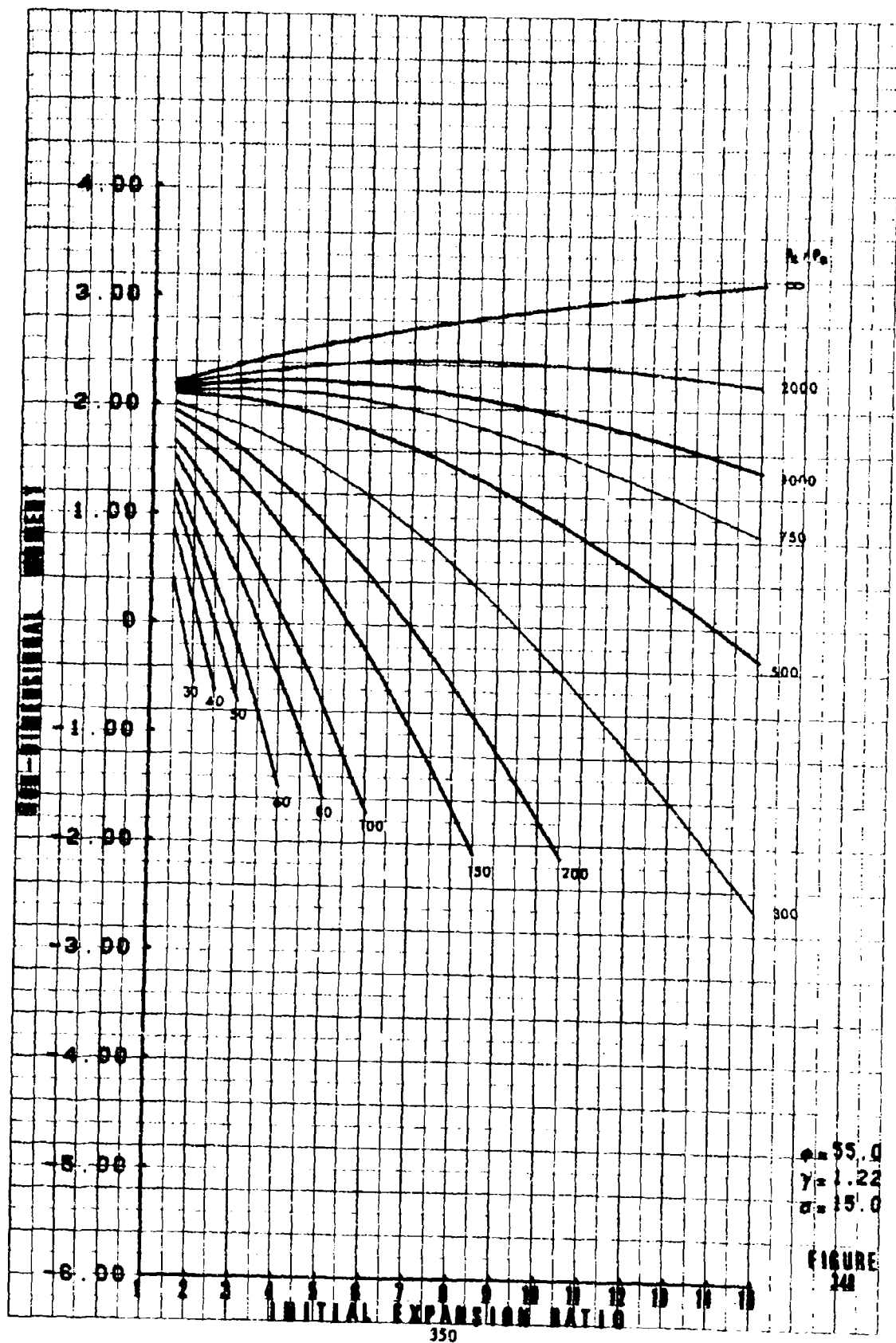
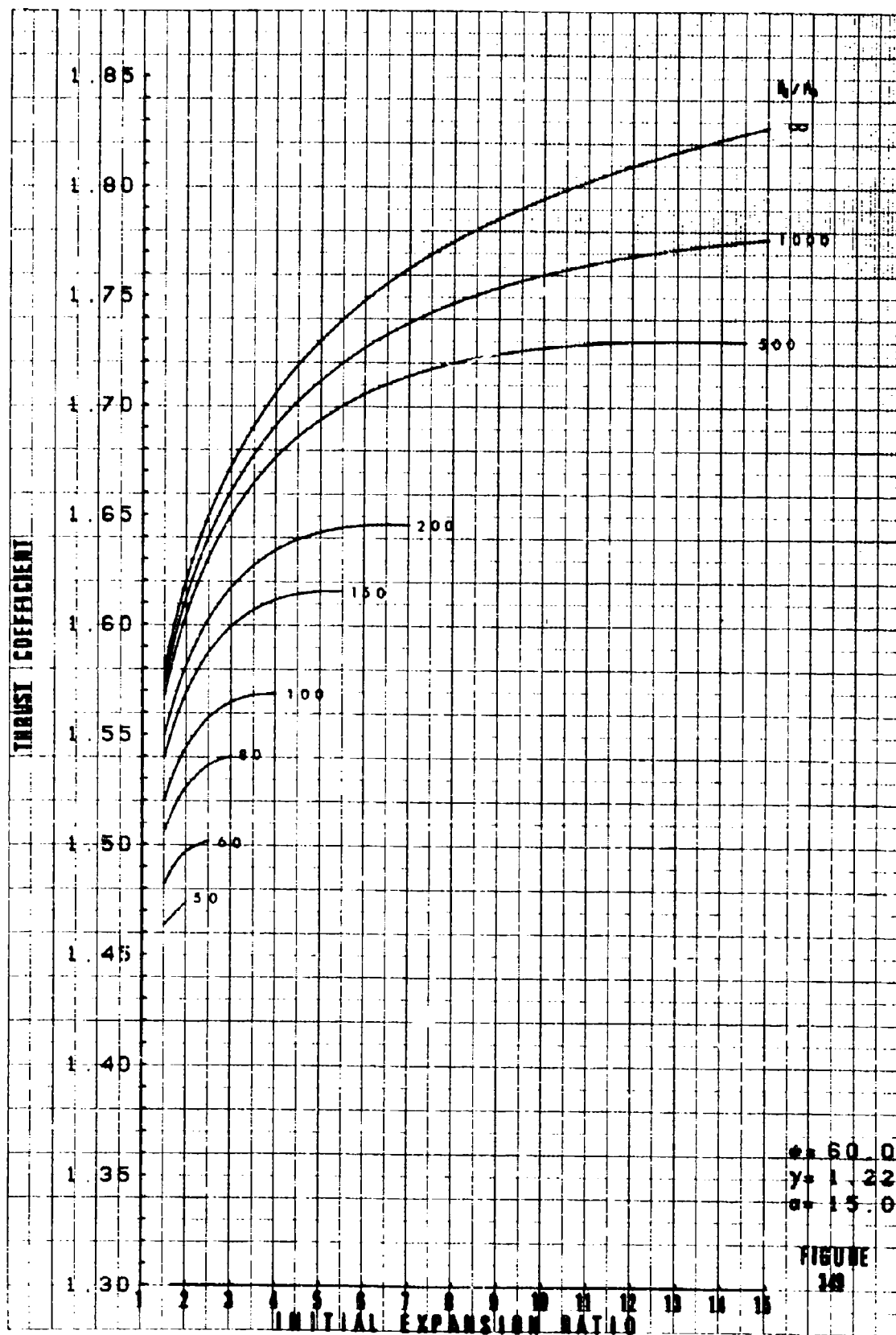
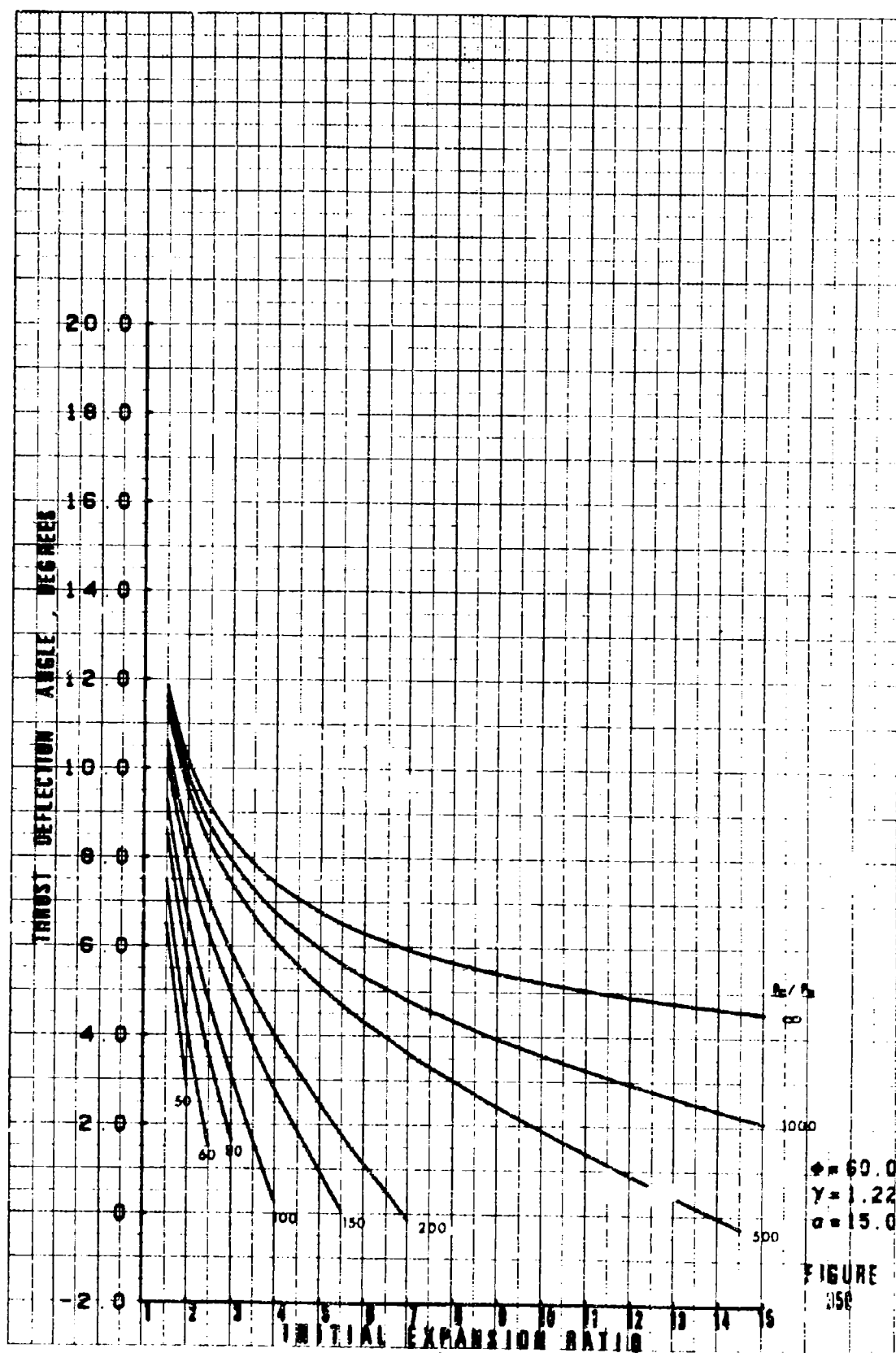
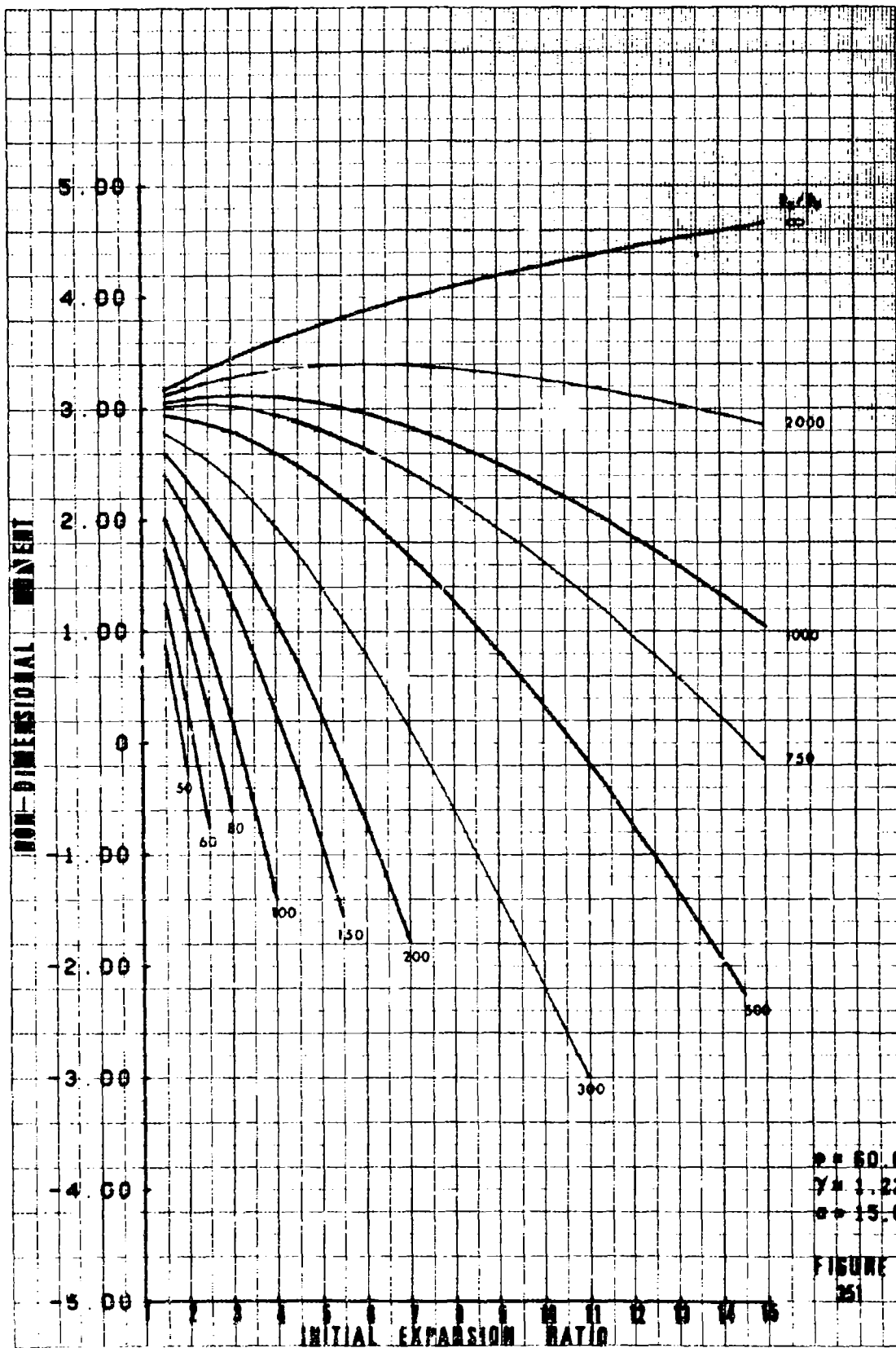


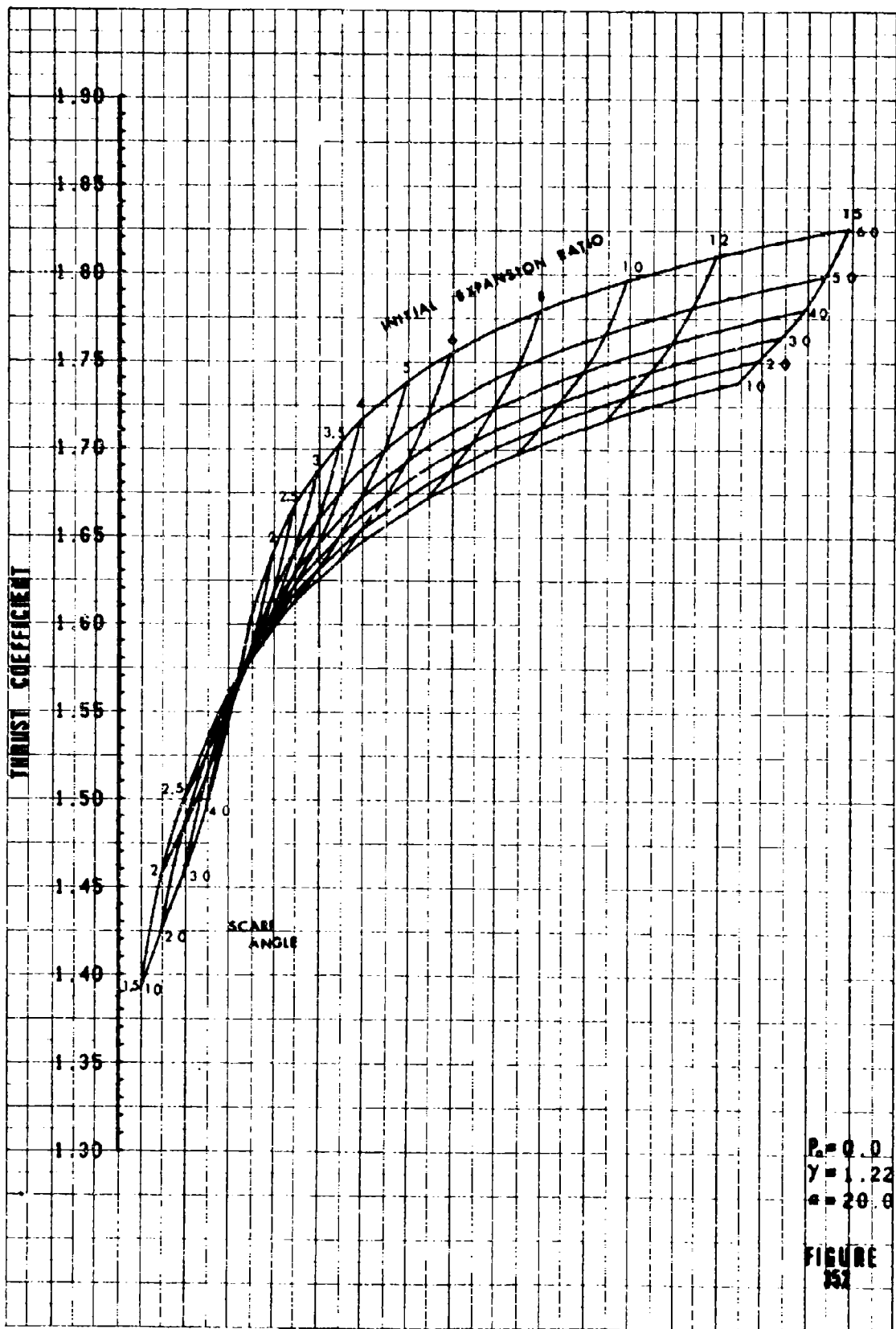
FIGURE 341

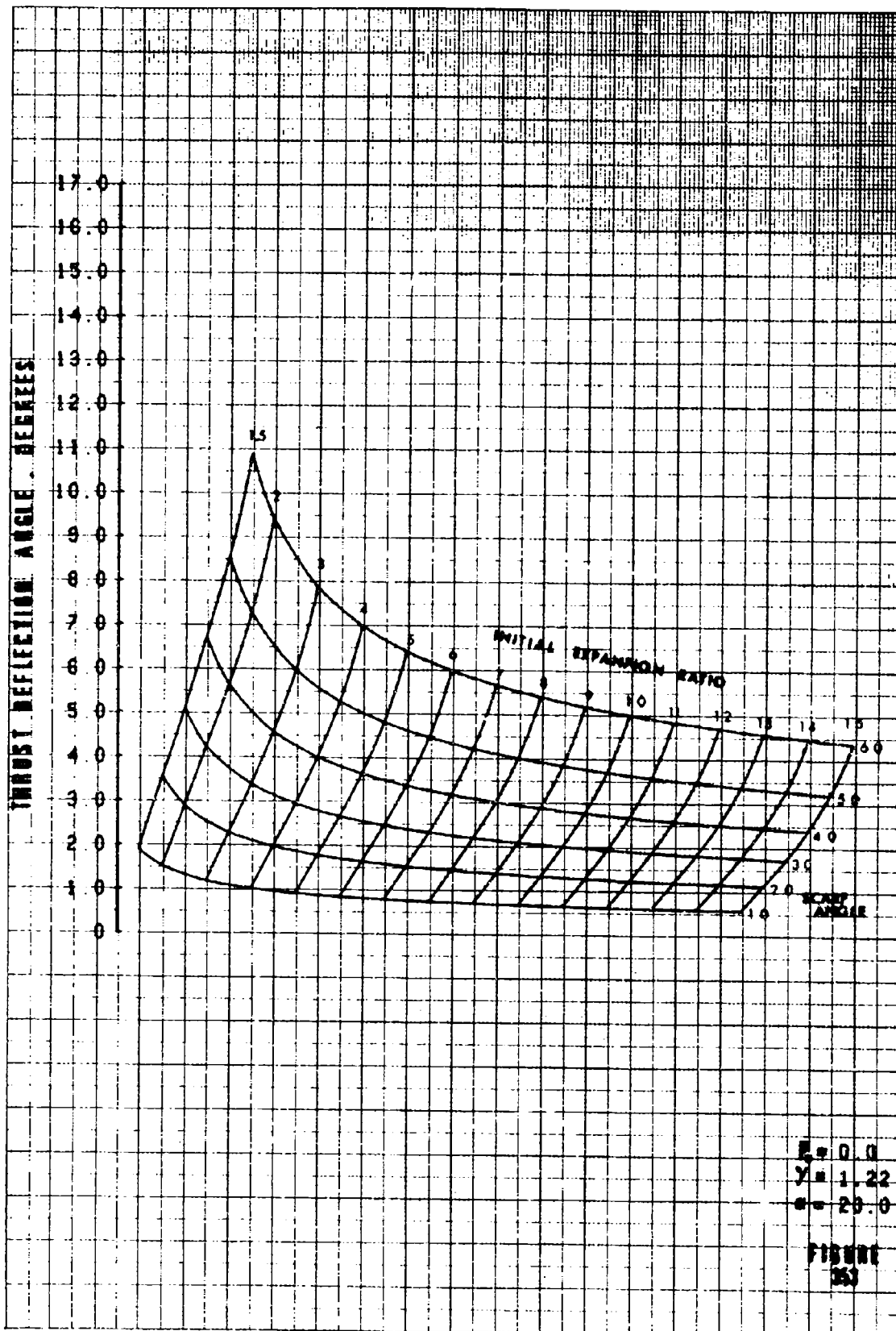


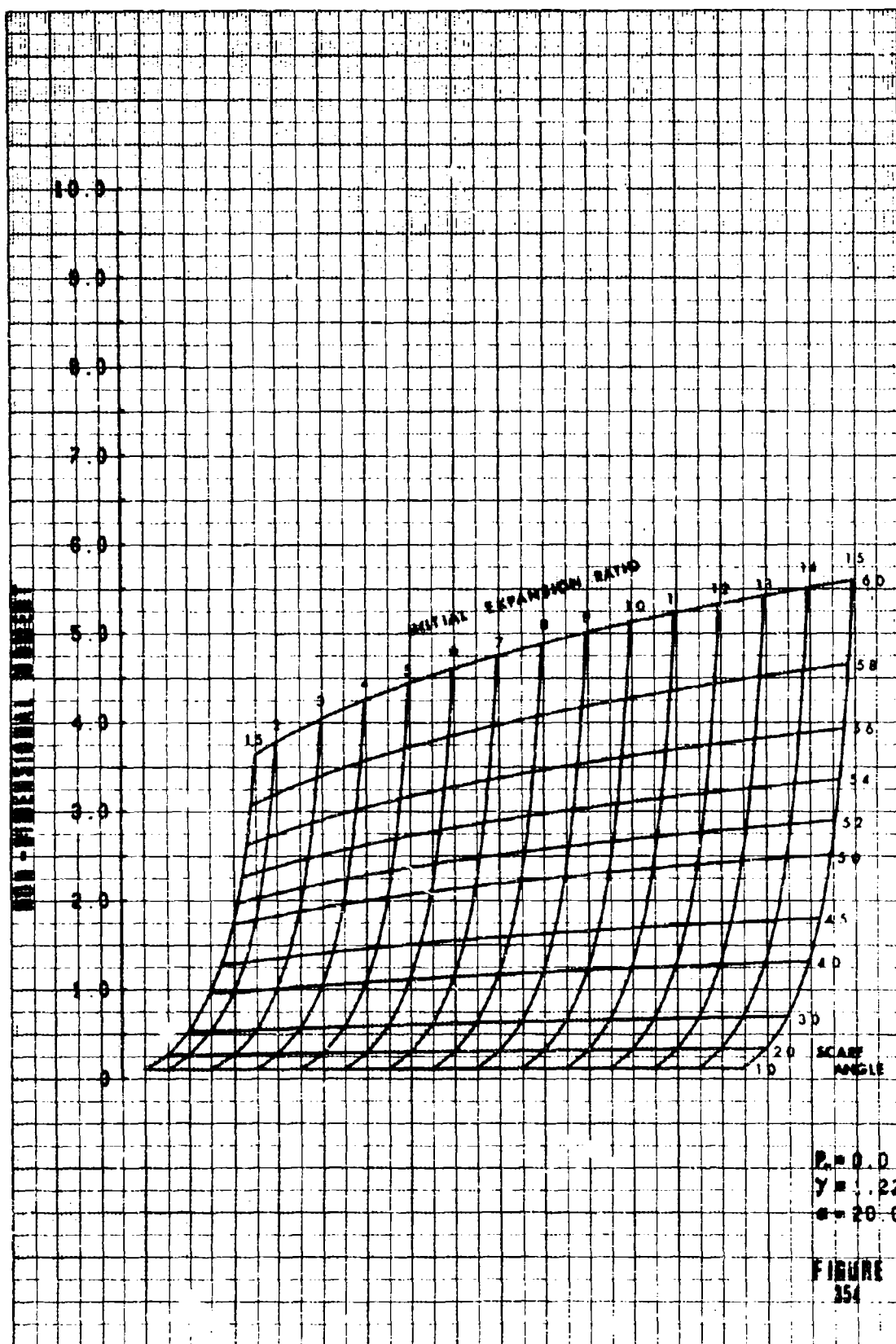












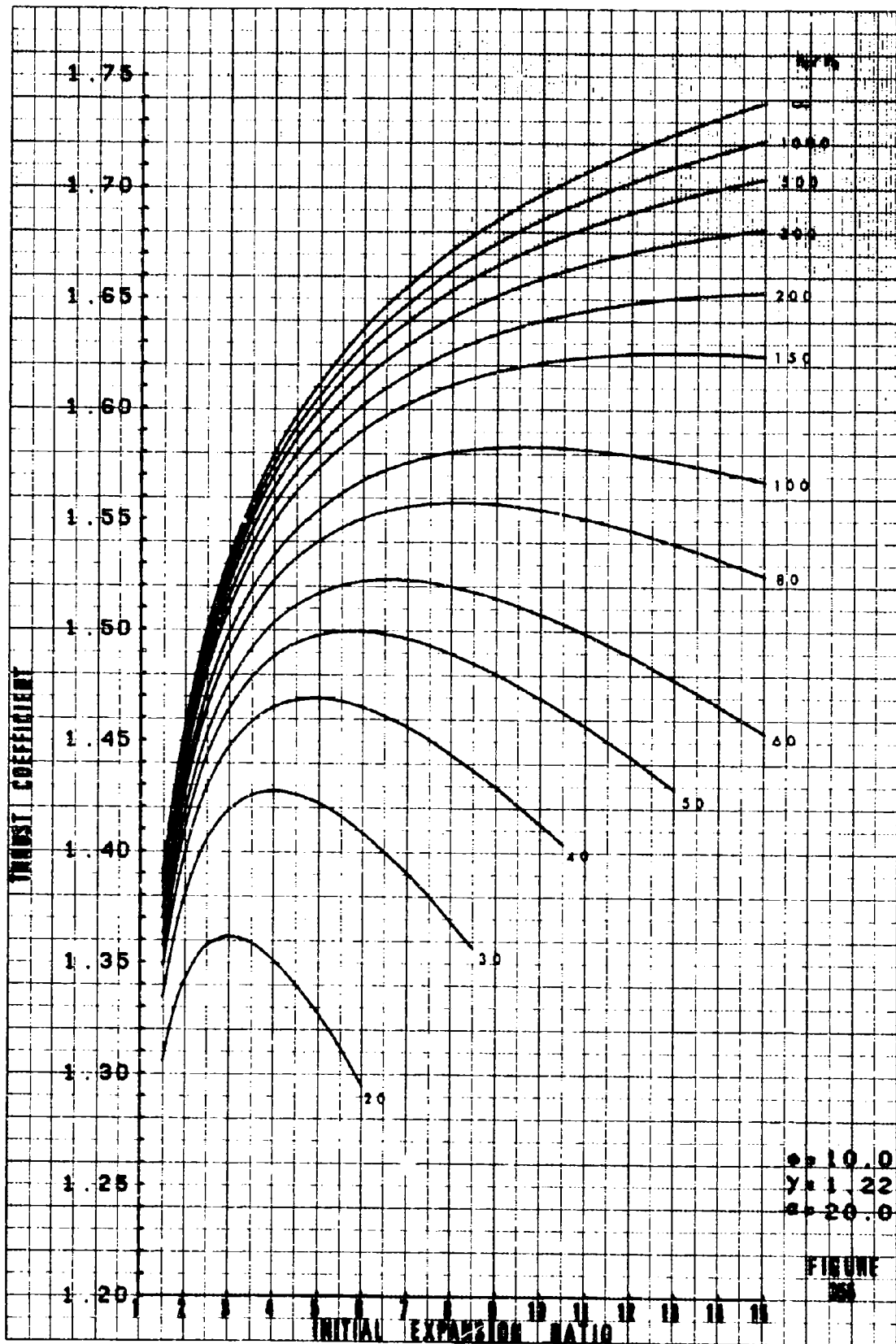
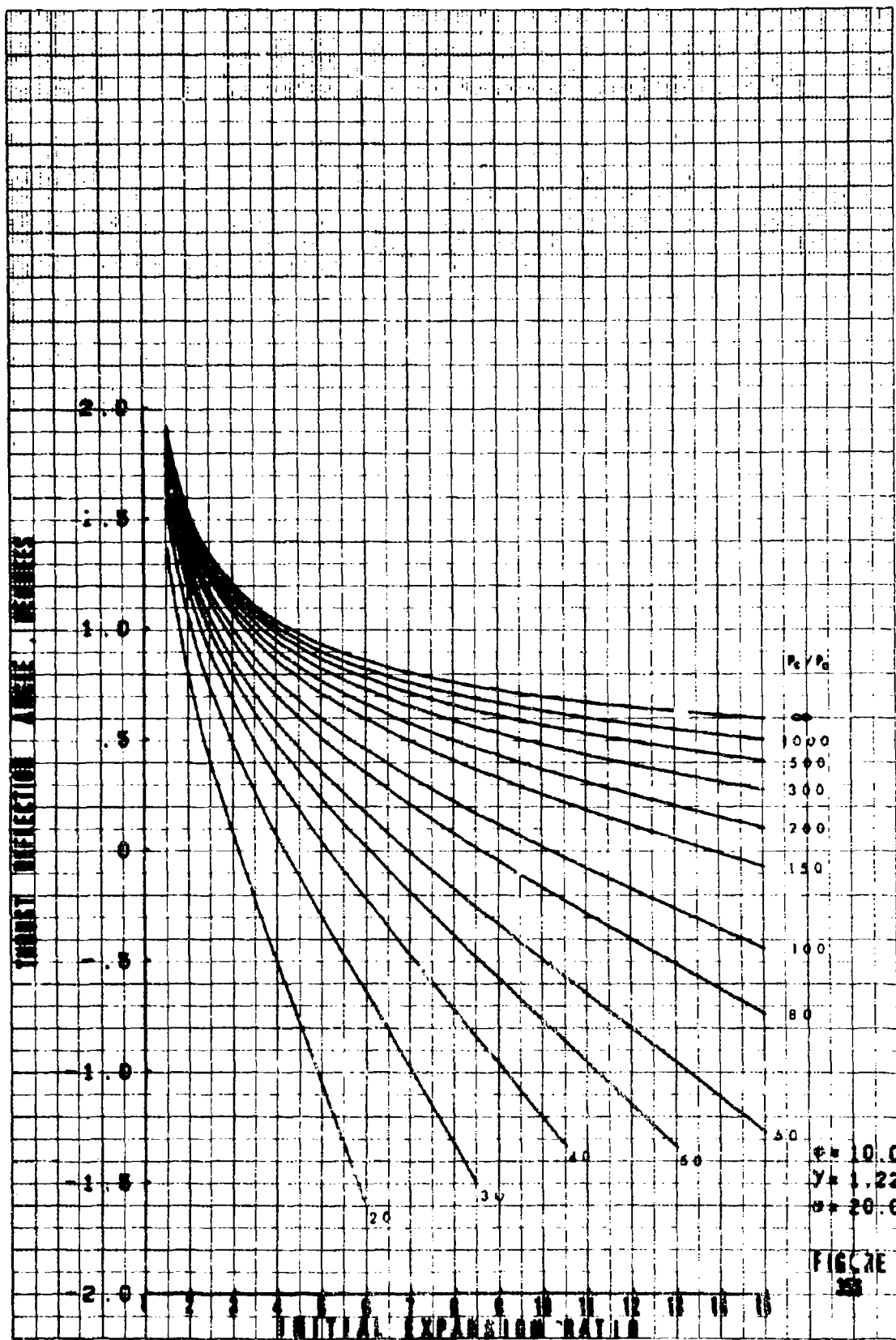
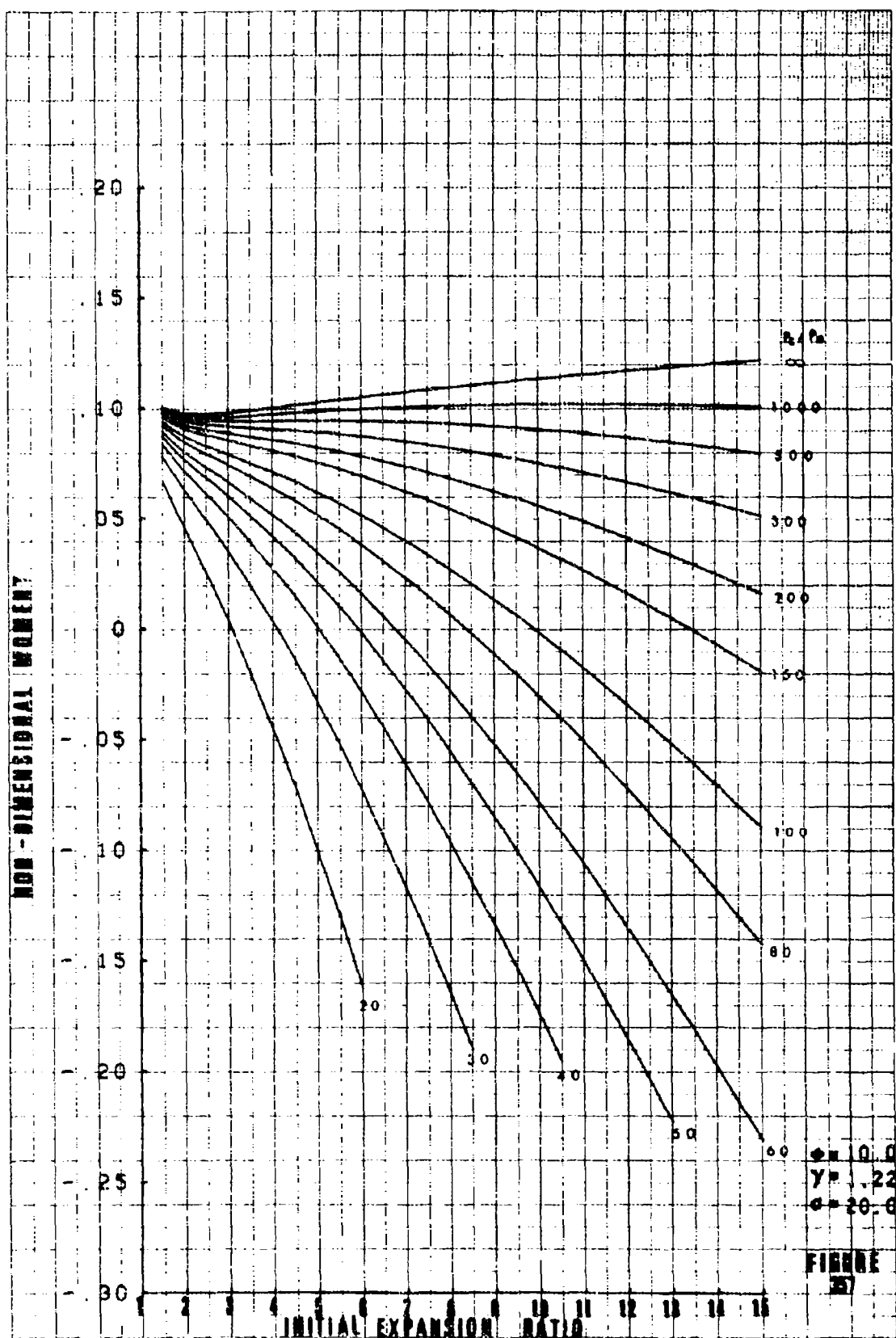
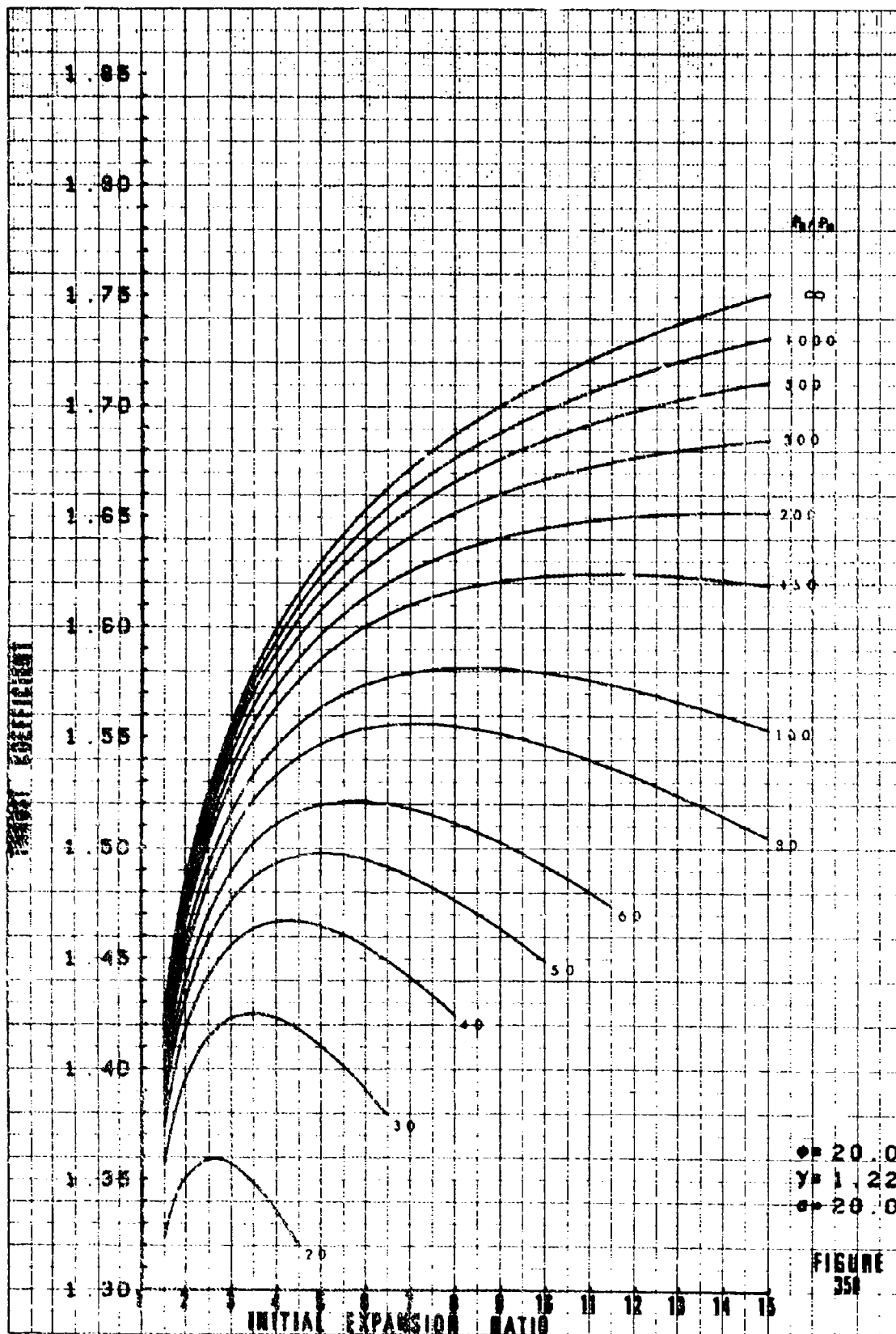


FIGURE 255







THRUST DEFLECTION ANGLE SERIES

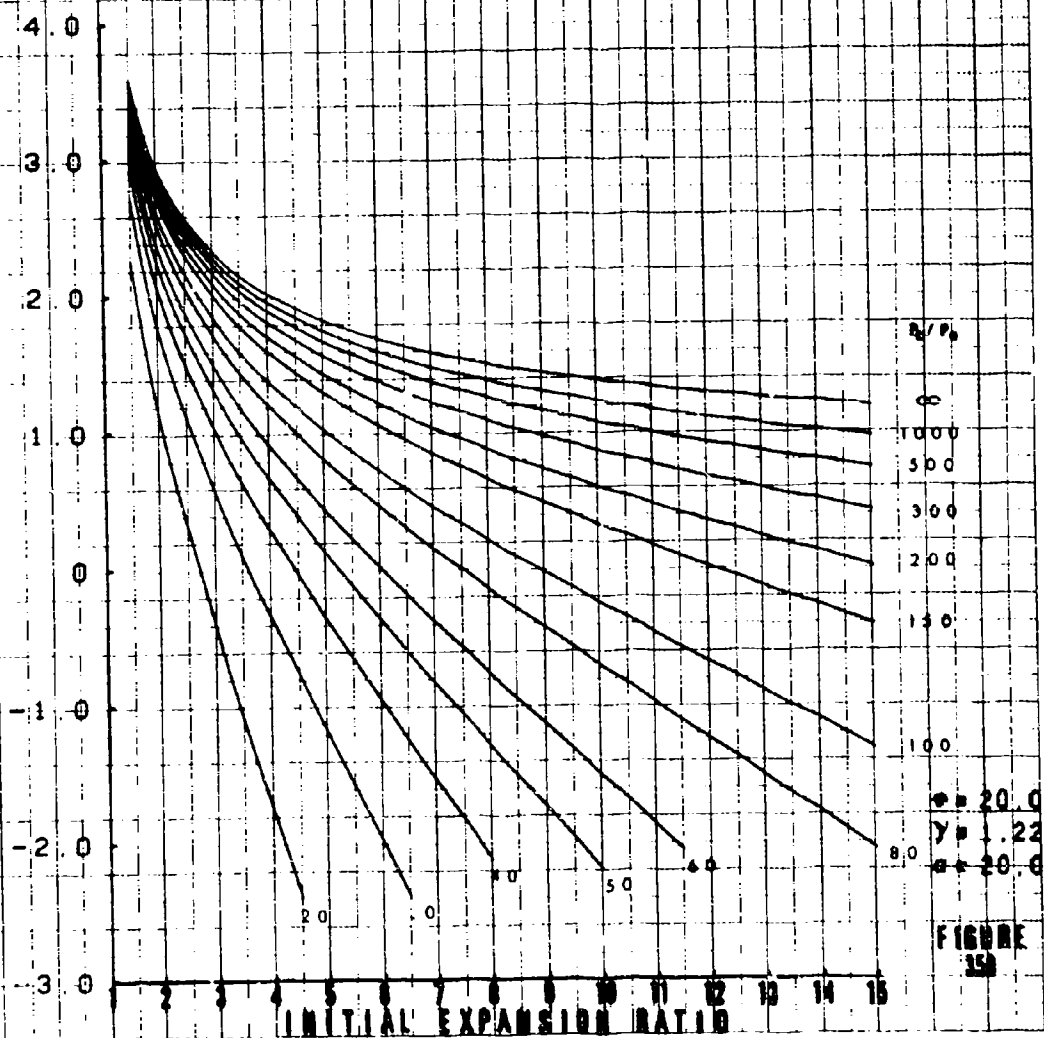
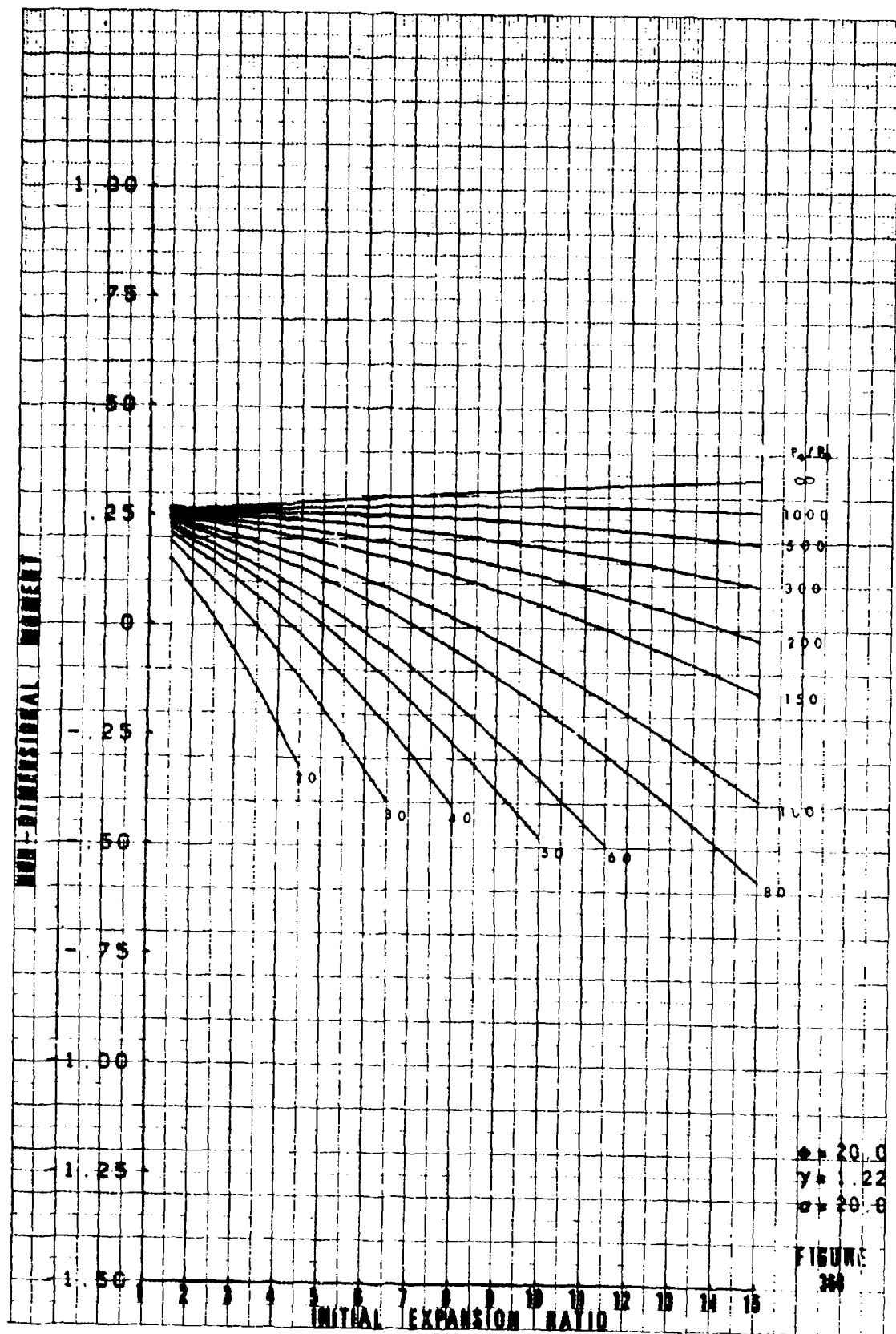
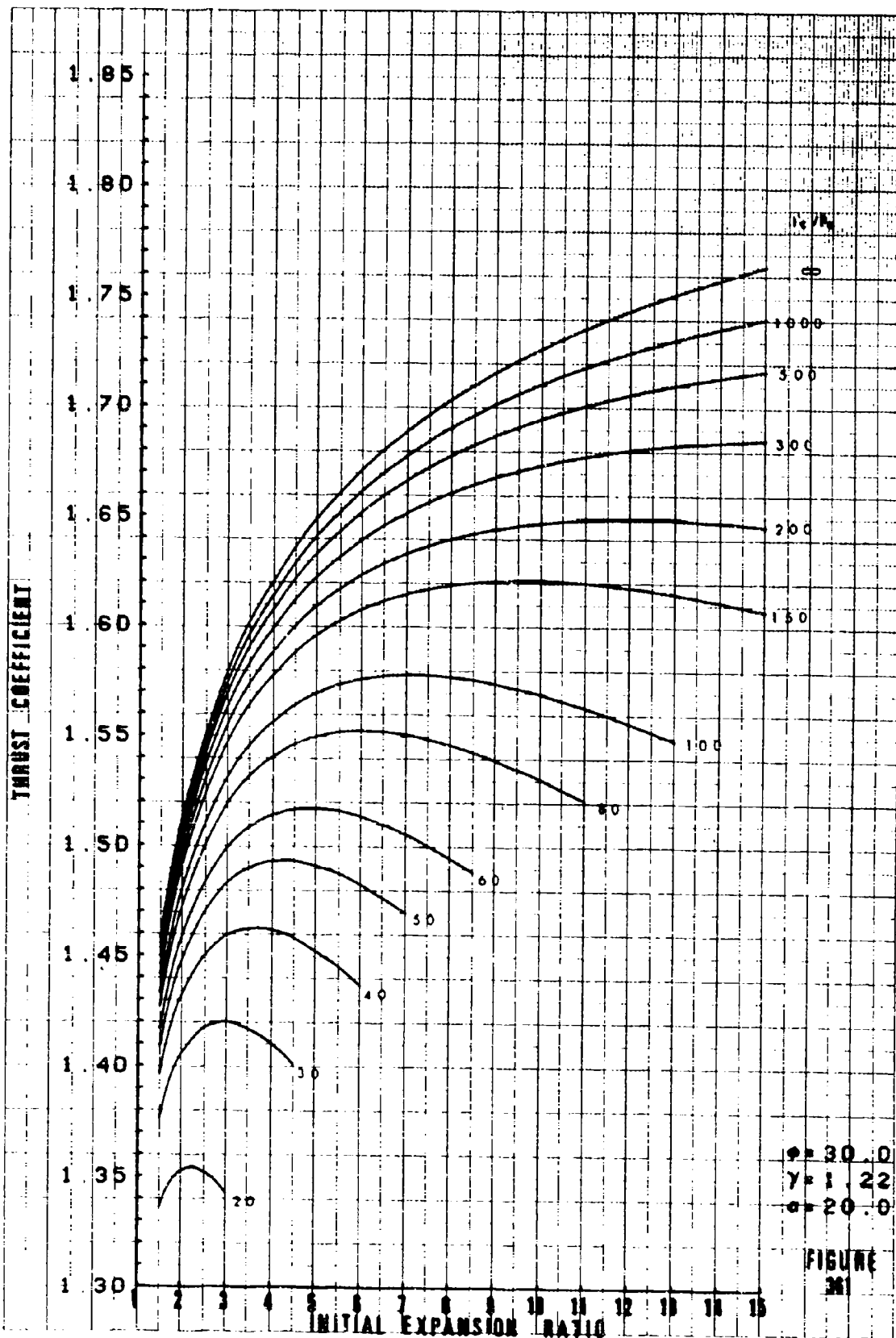
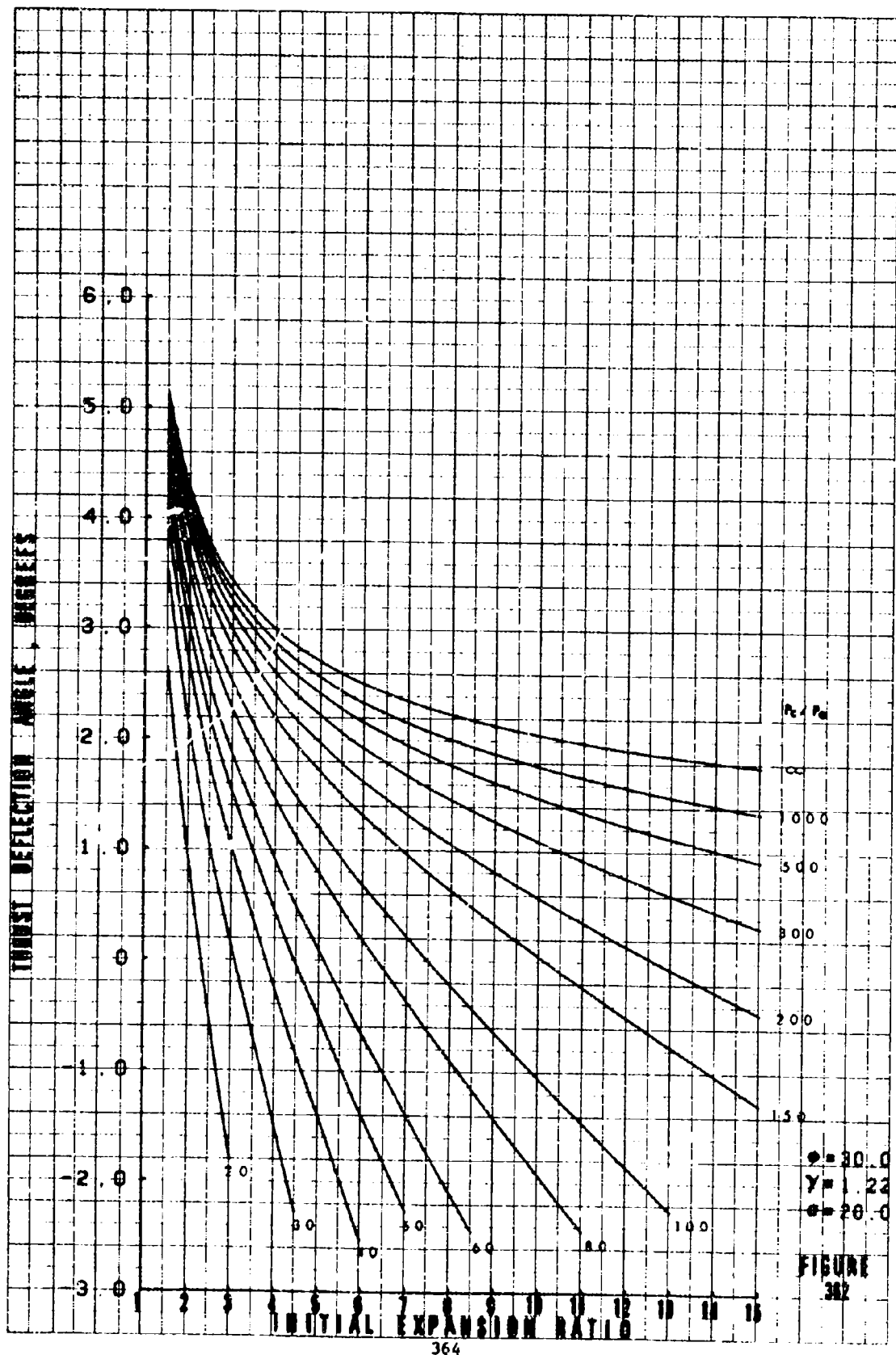
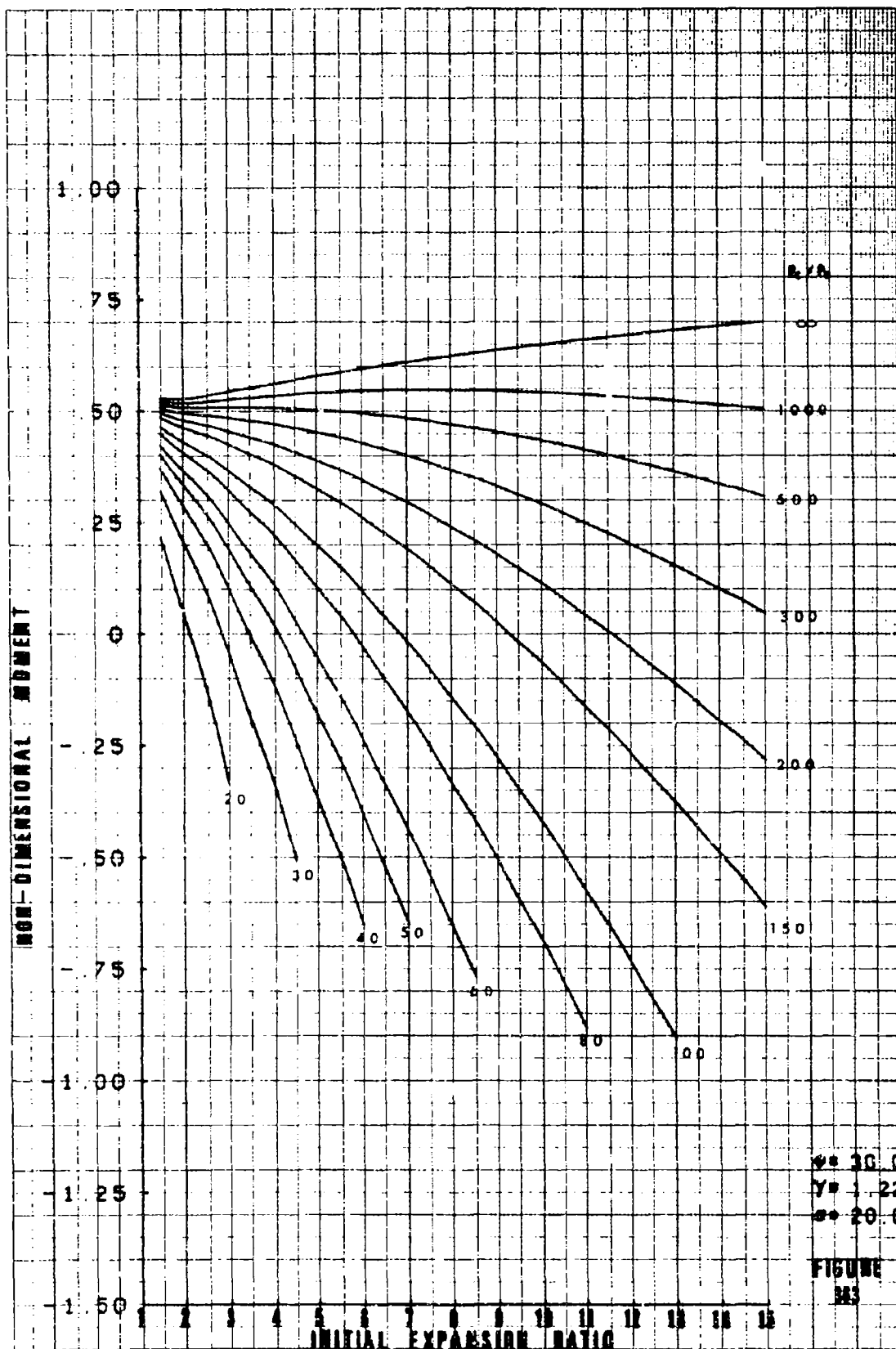


FIGURE 15



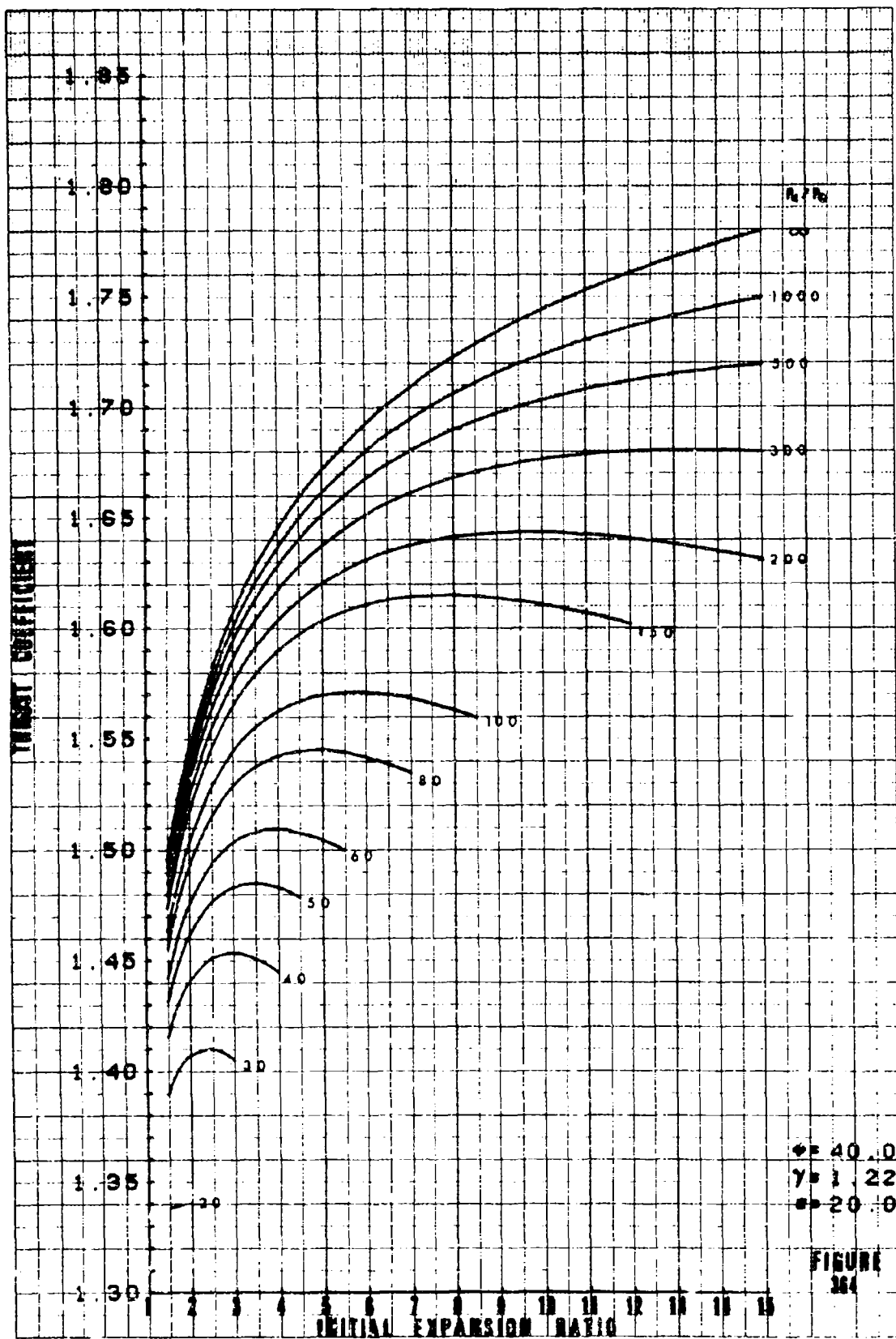






$\mu = 10.0$
 $\gamma = 1.22$
 $\sigma = 20.0$

FIGURE 363



$\gamma = 40.0$
 $\gamma = 1.22$
 $\gamma = 20.0$

FIGURE 364

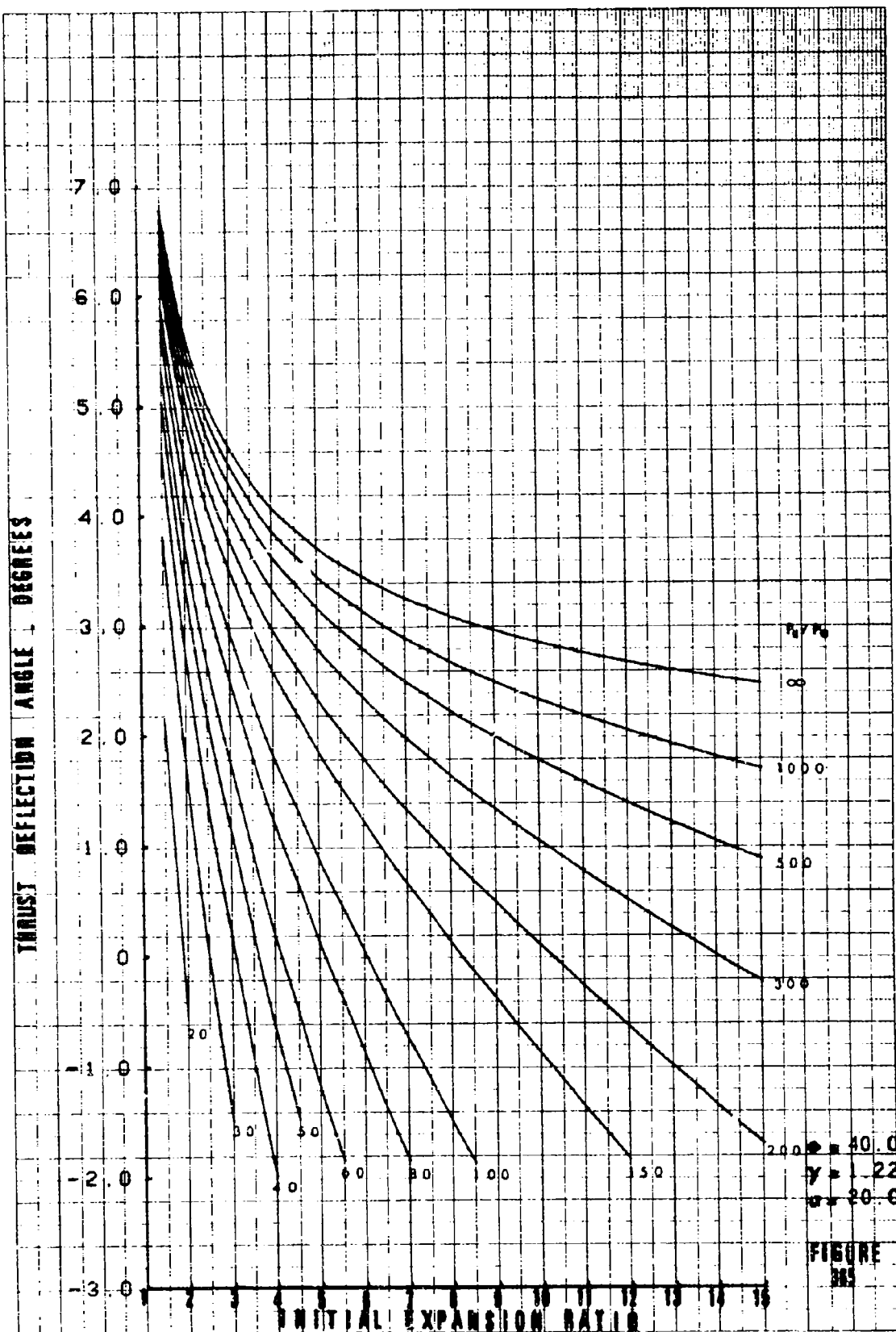
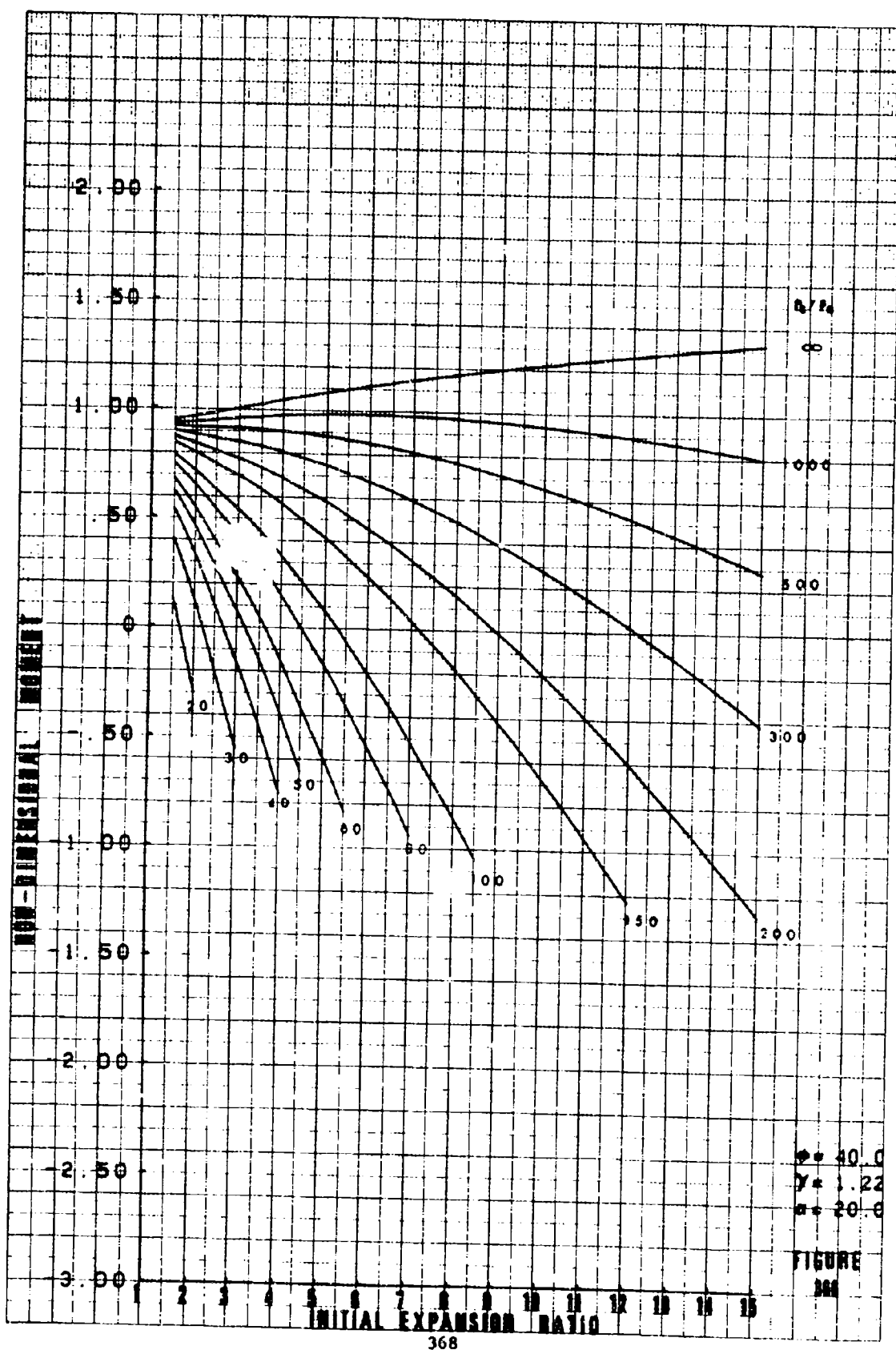
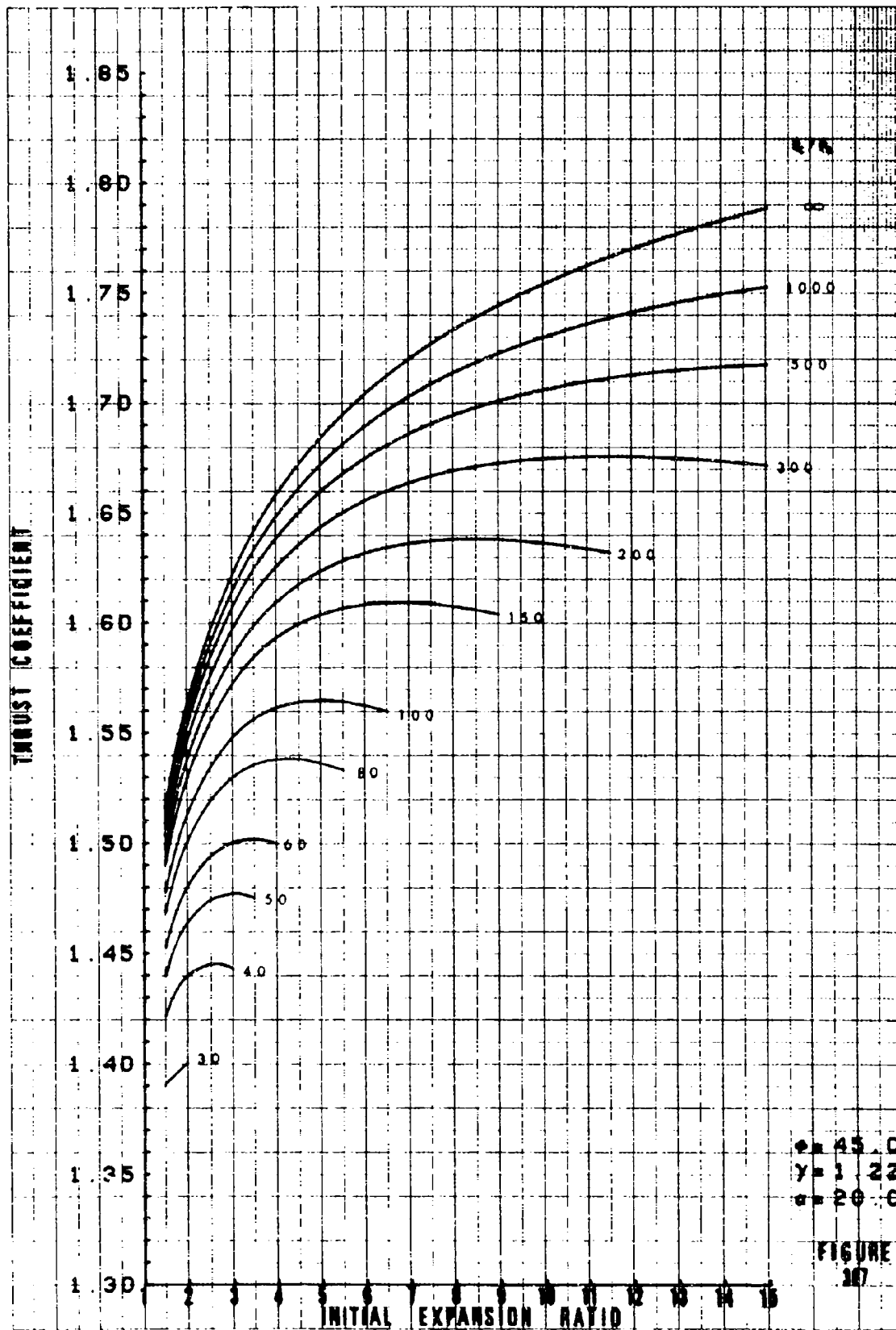


FIGURE 365





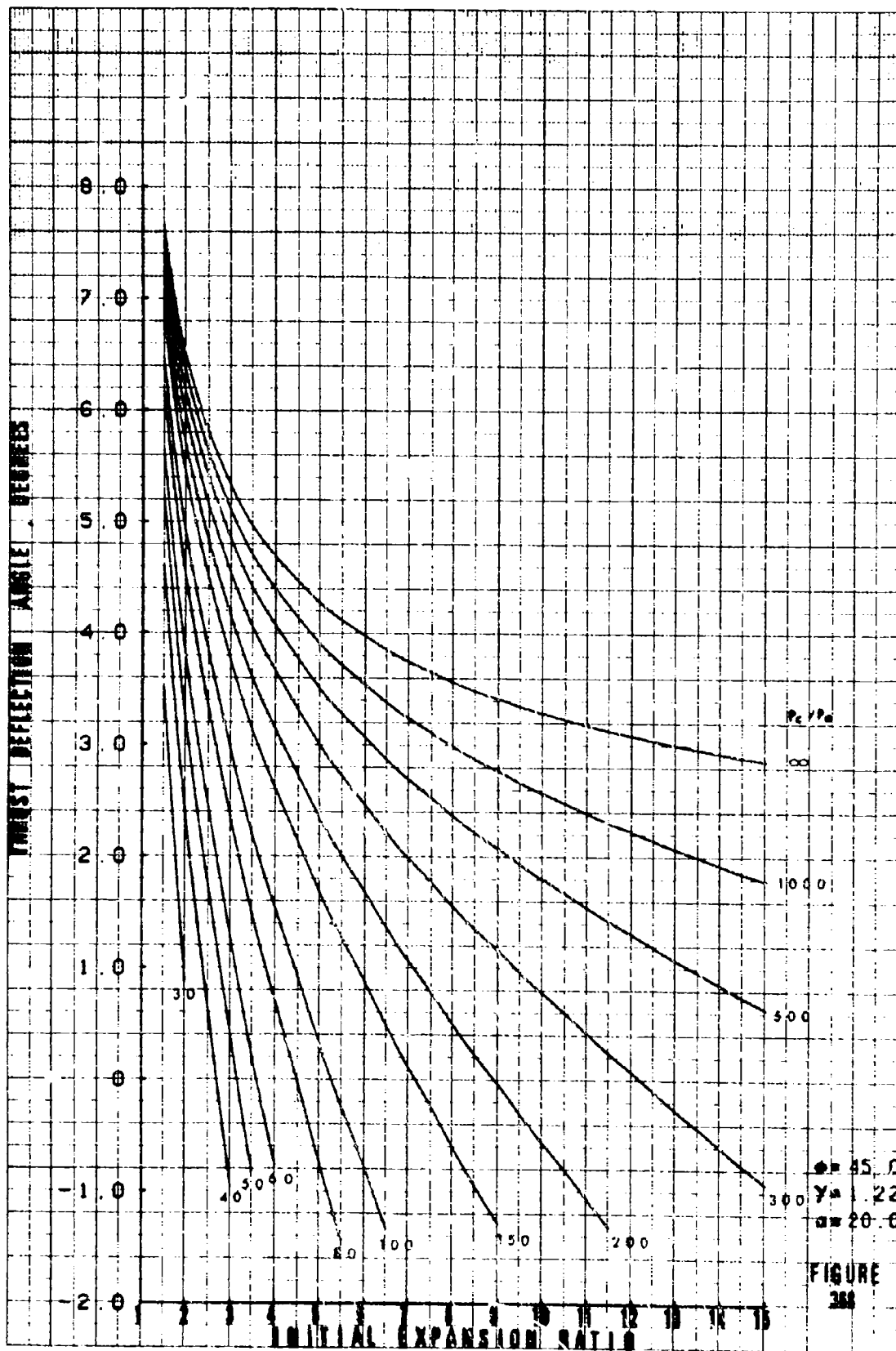
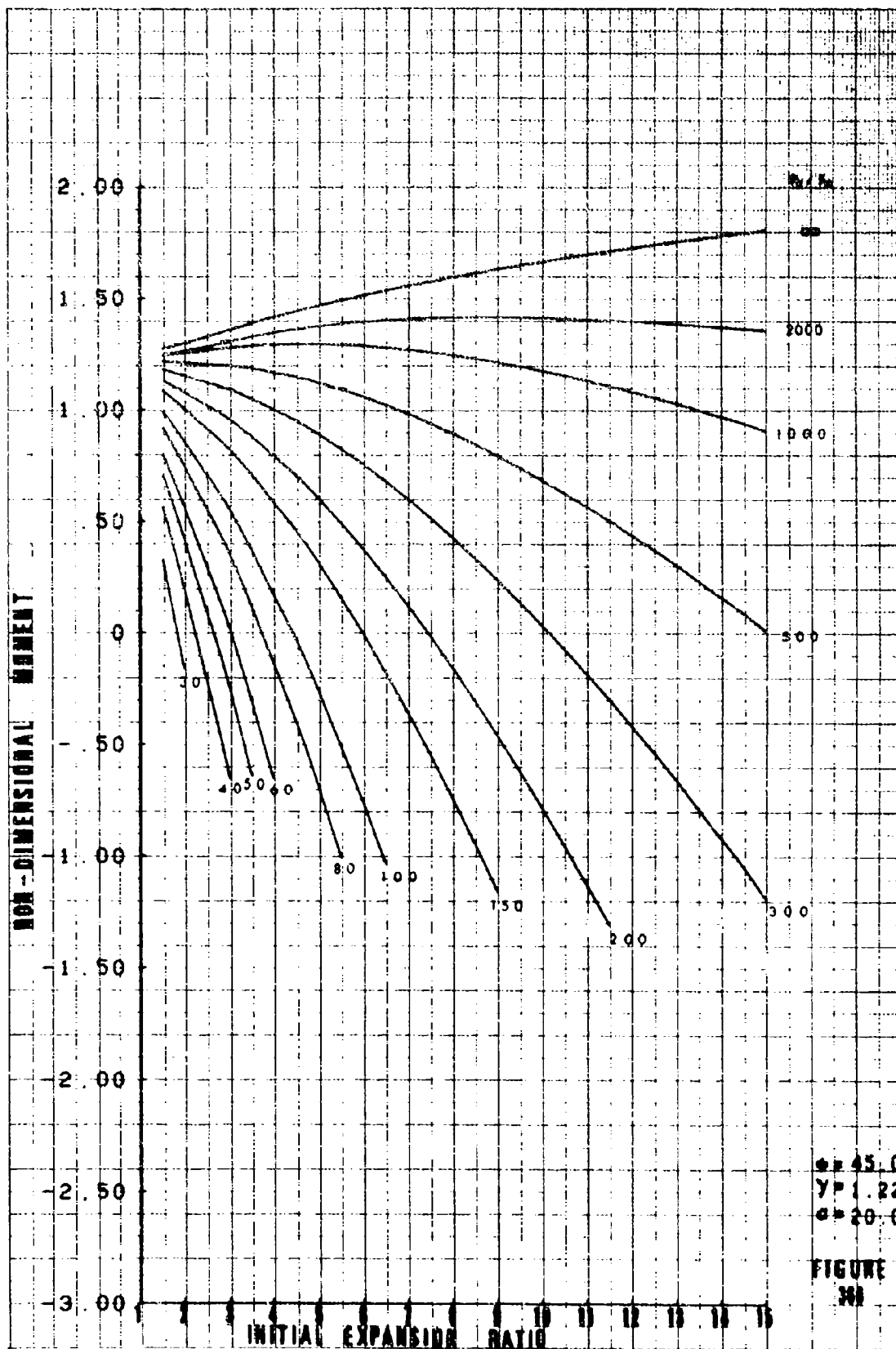
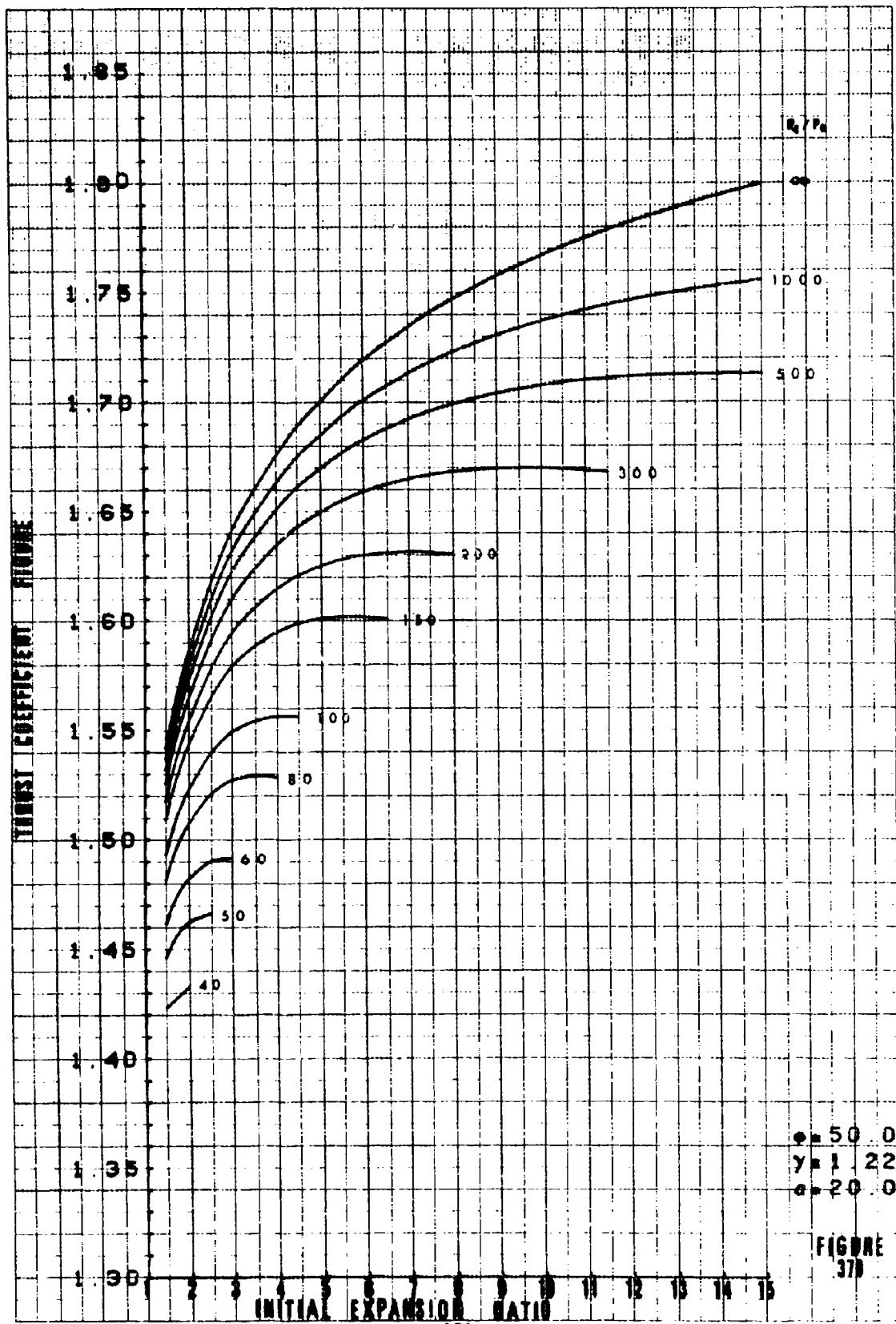


FIGURE
368





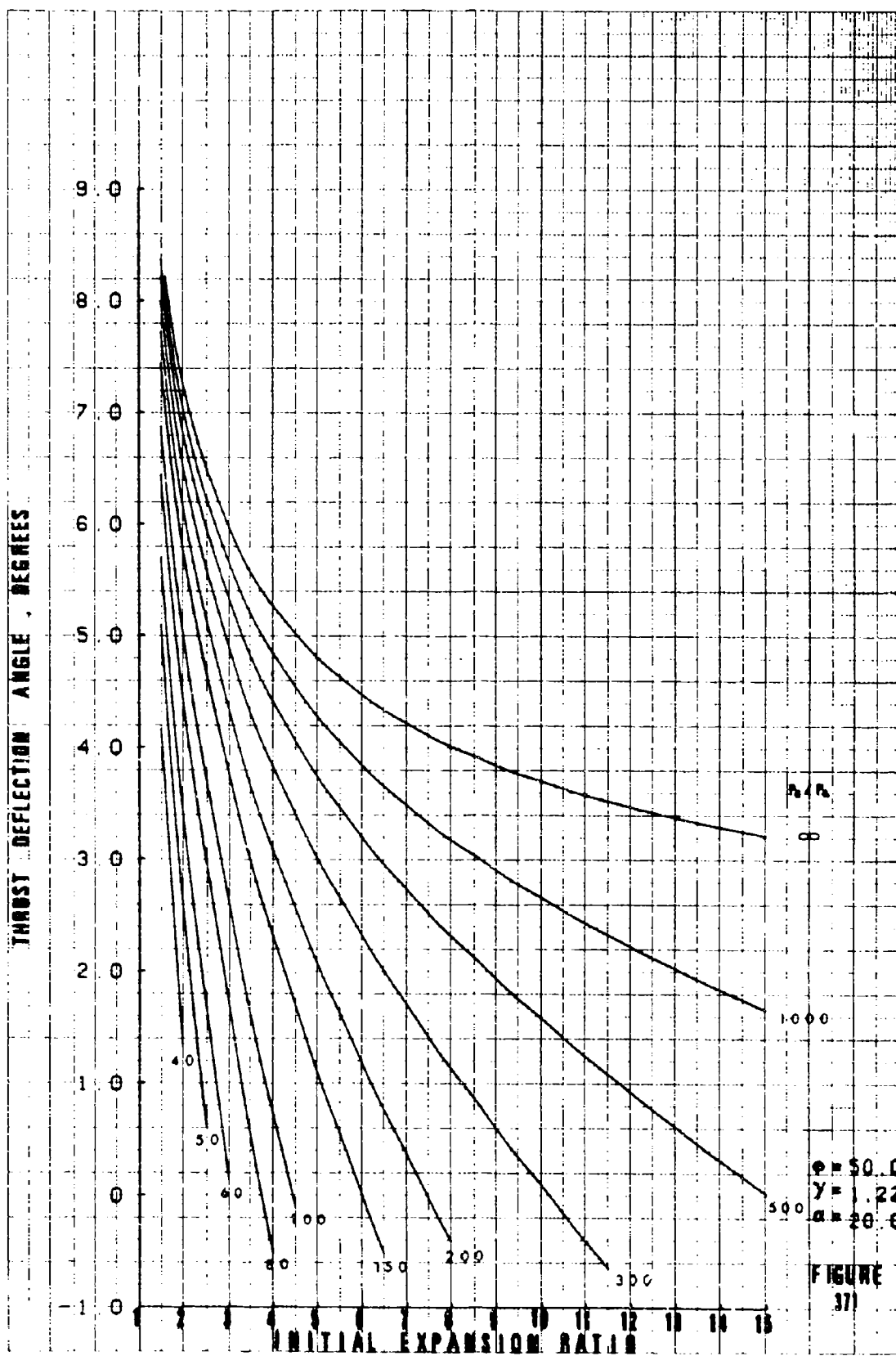
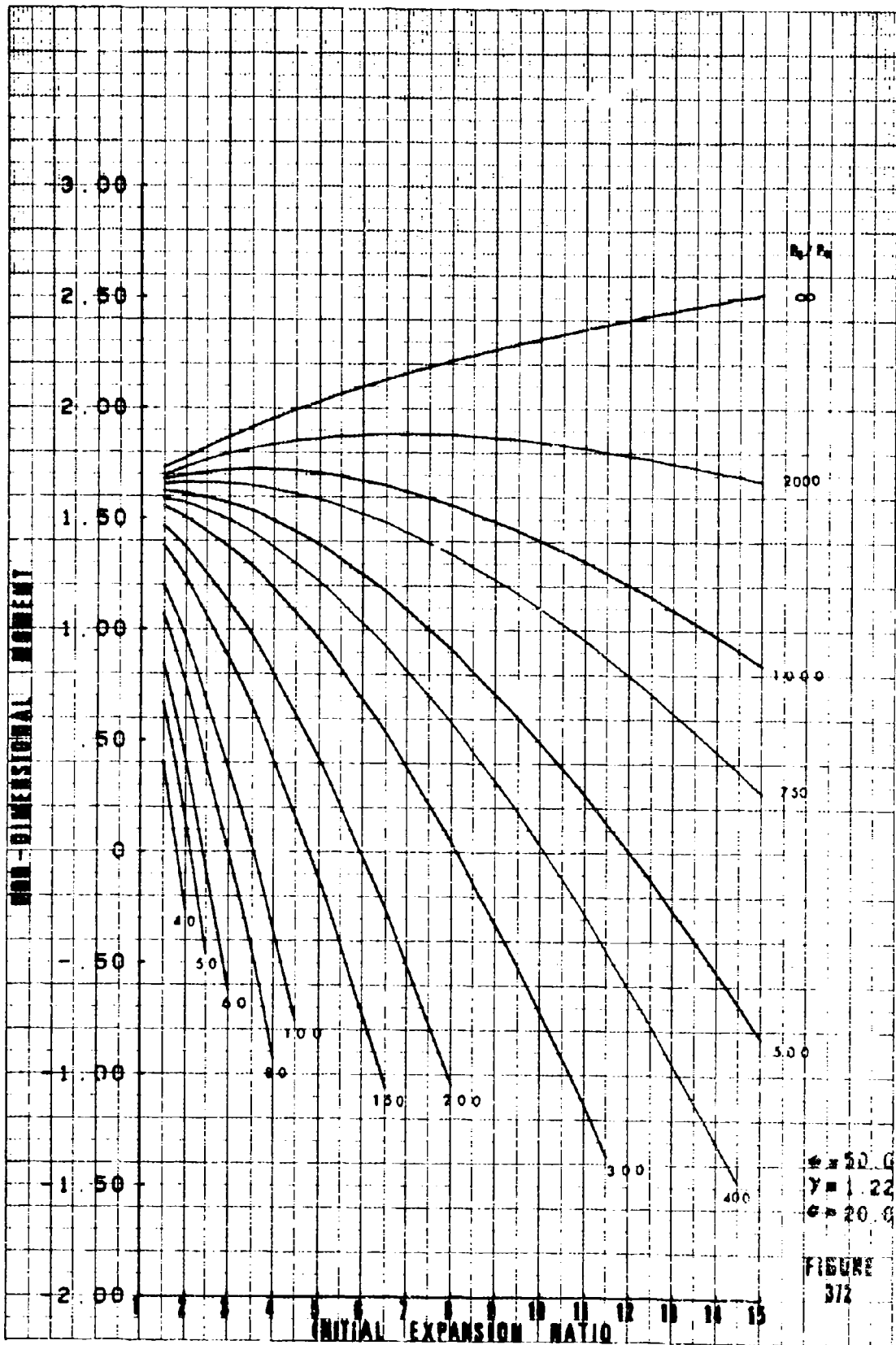
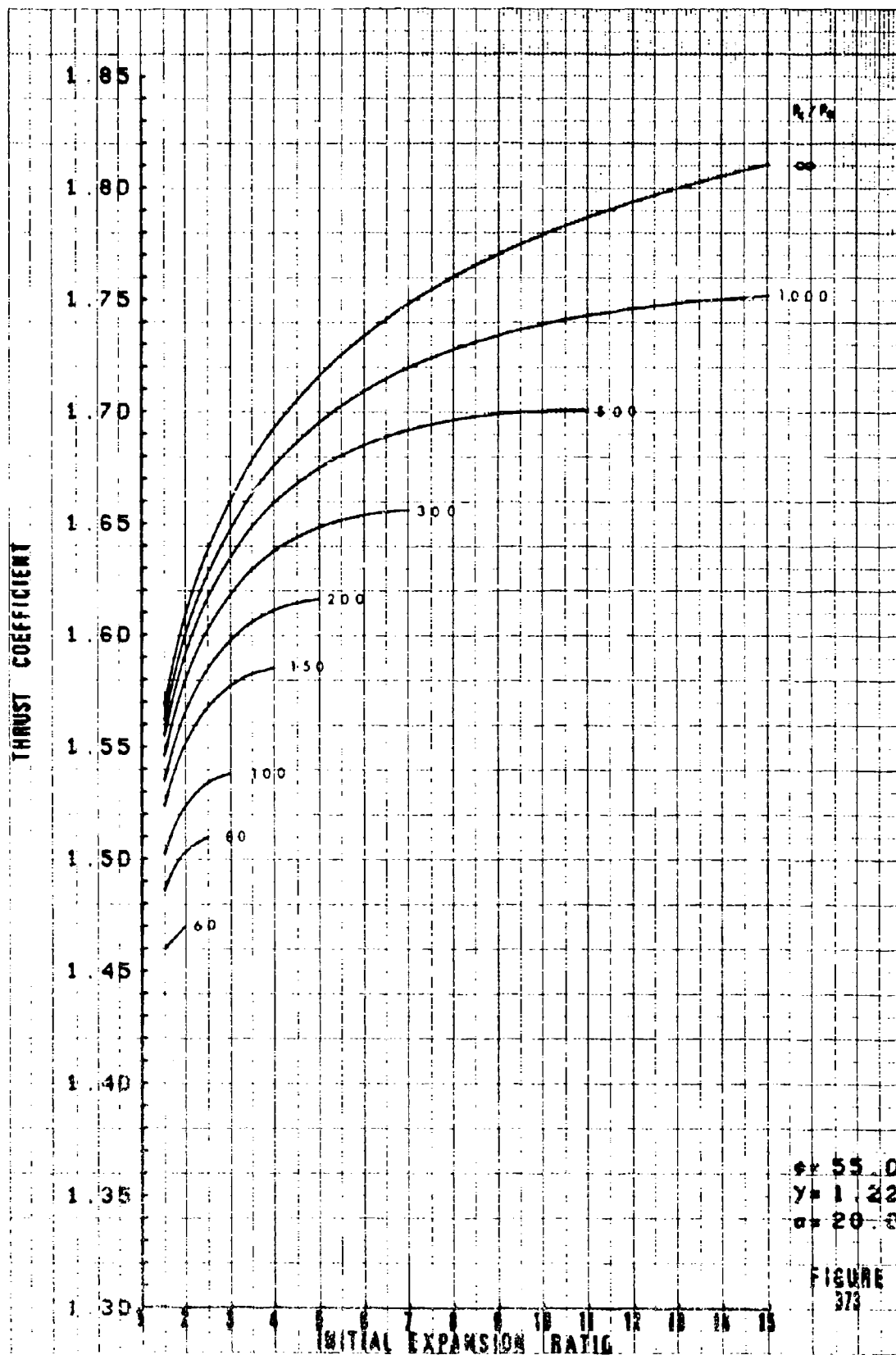


FIGURE 37





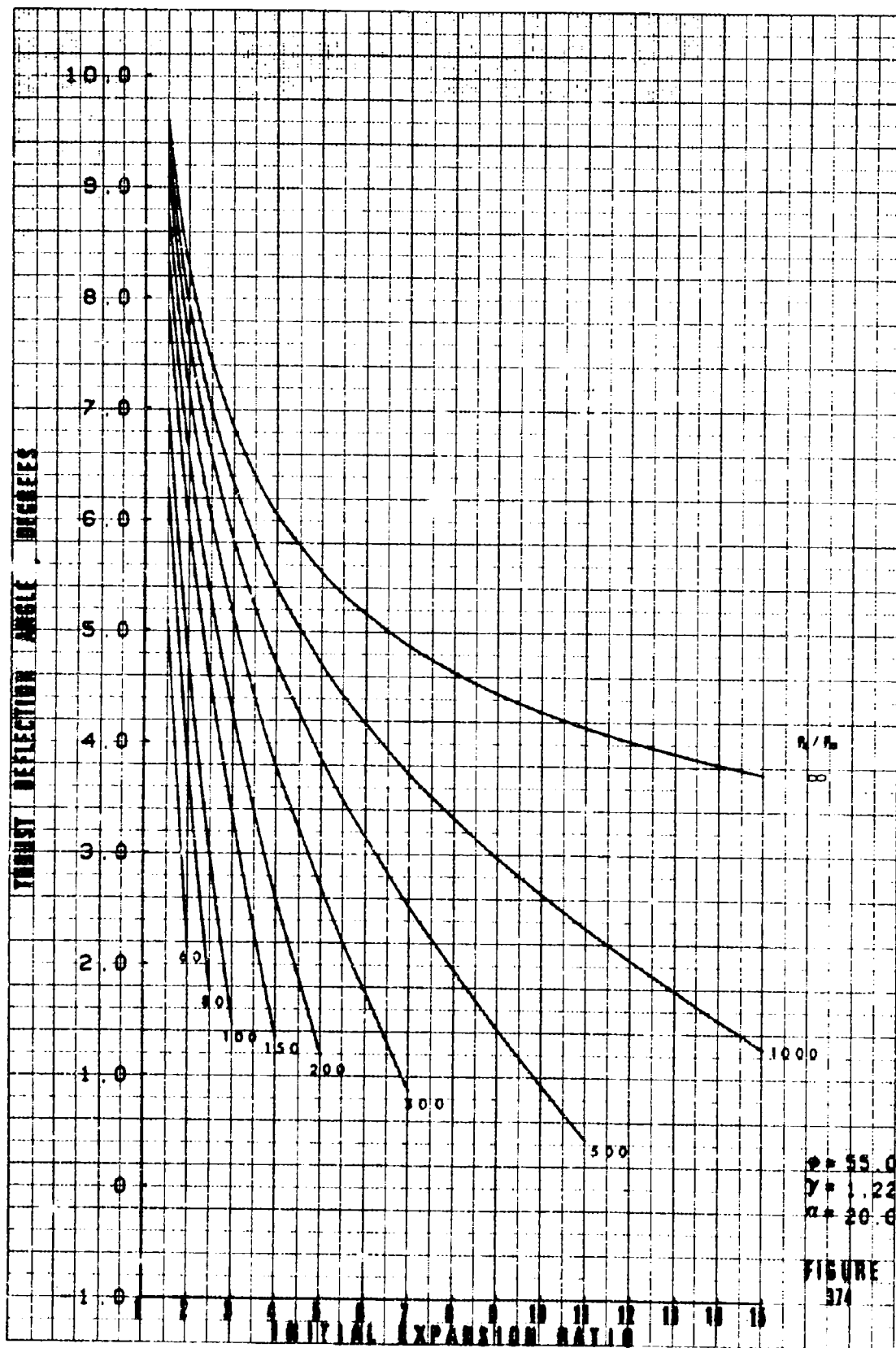
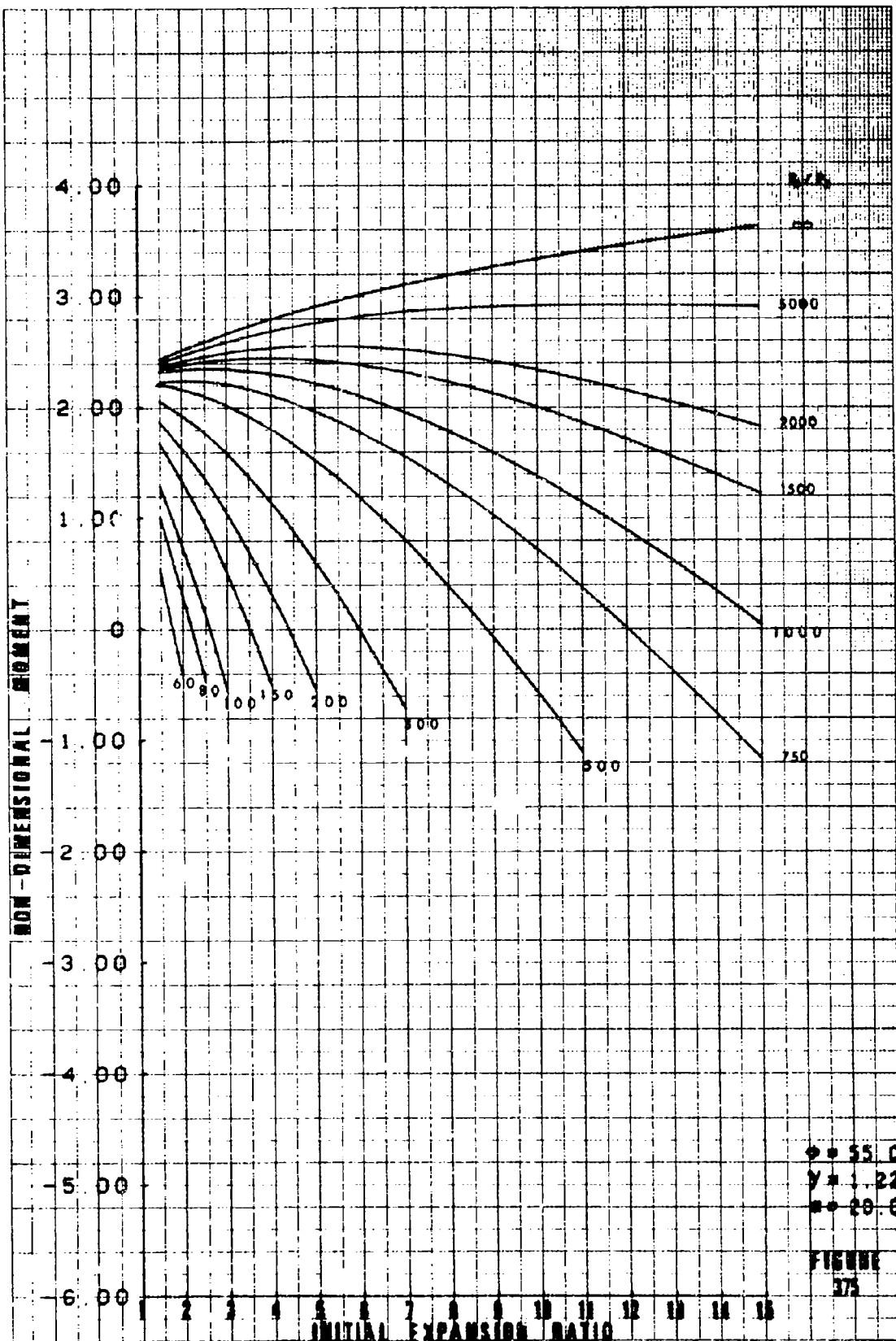
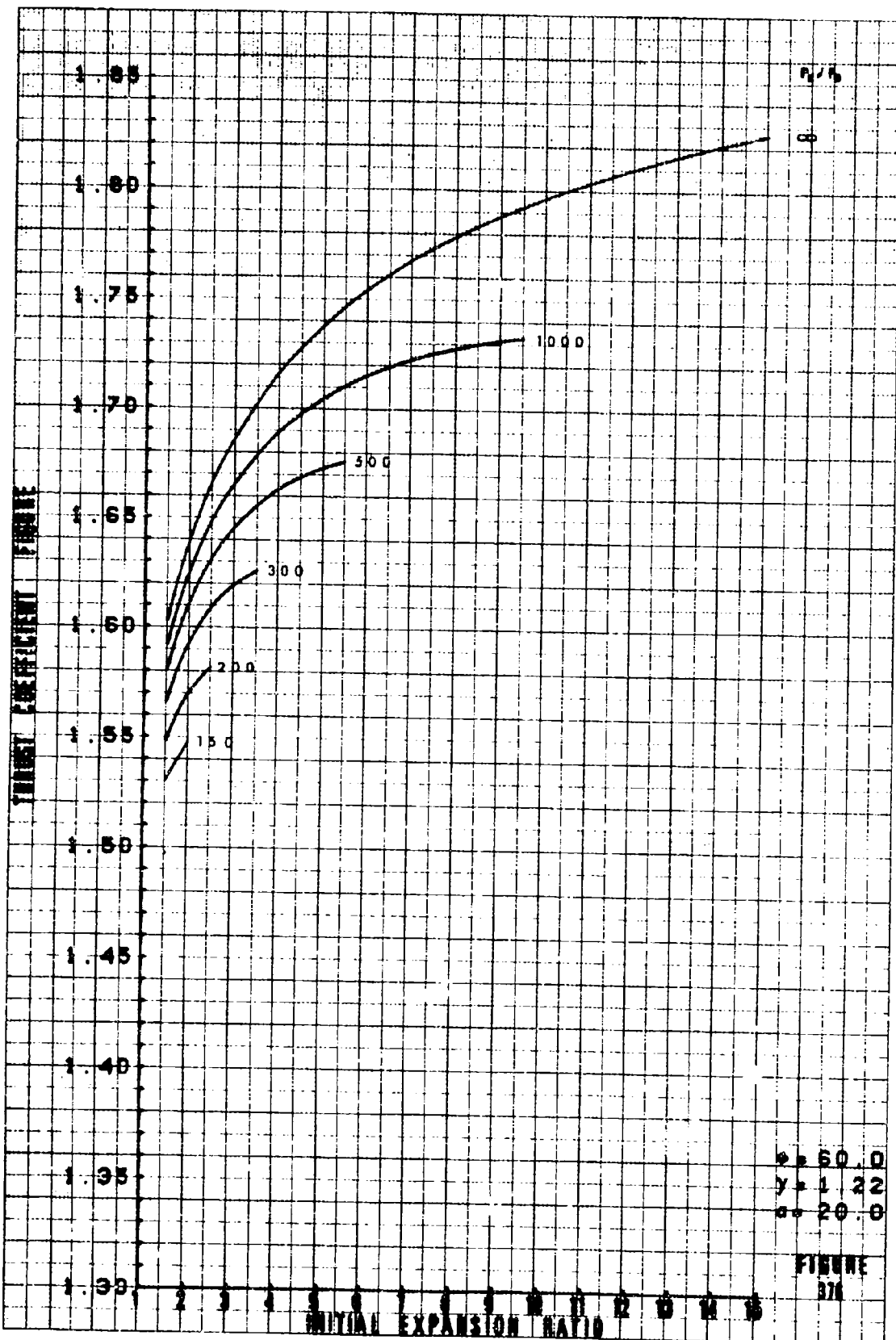
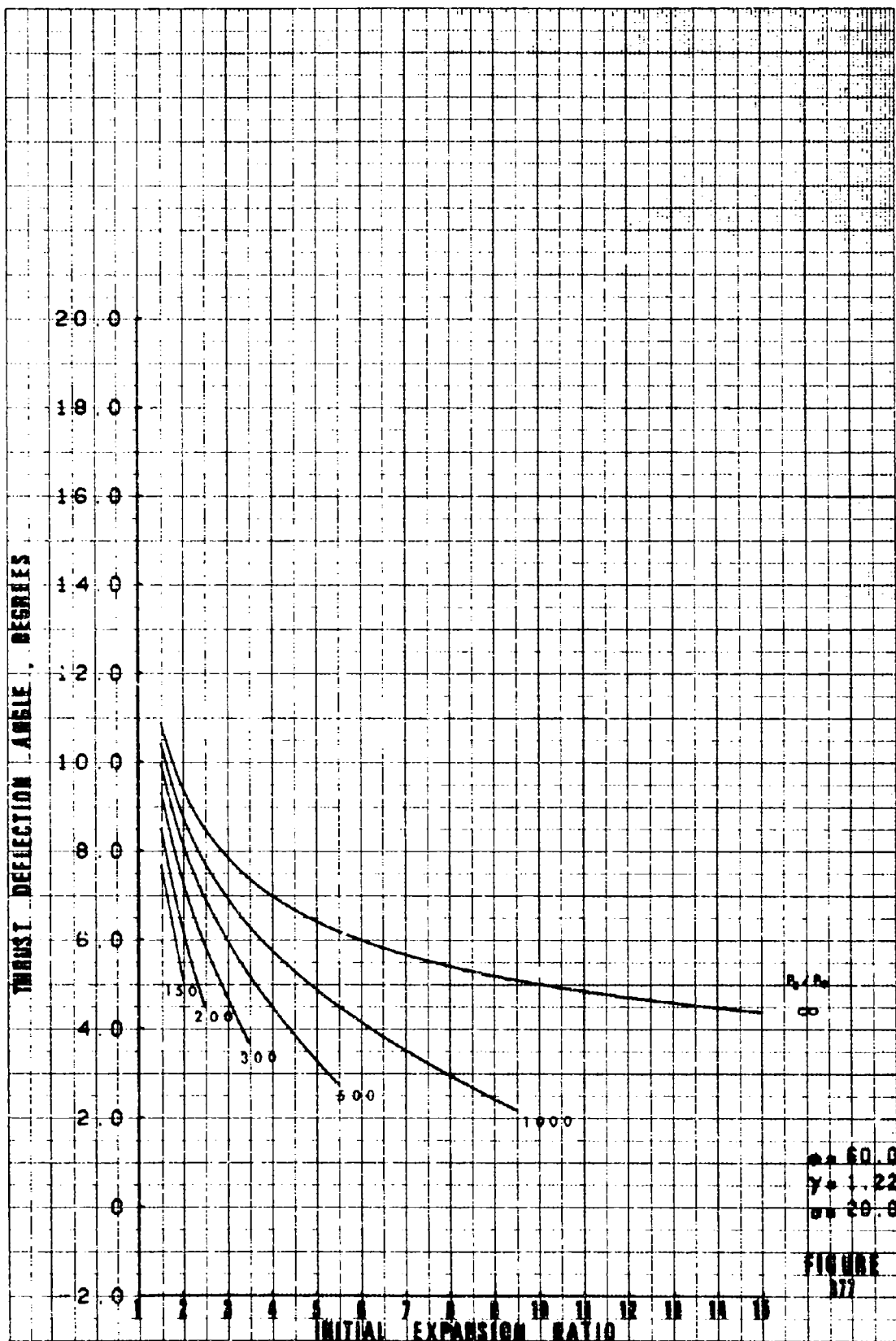
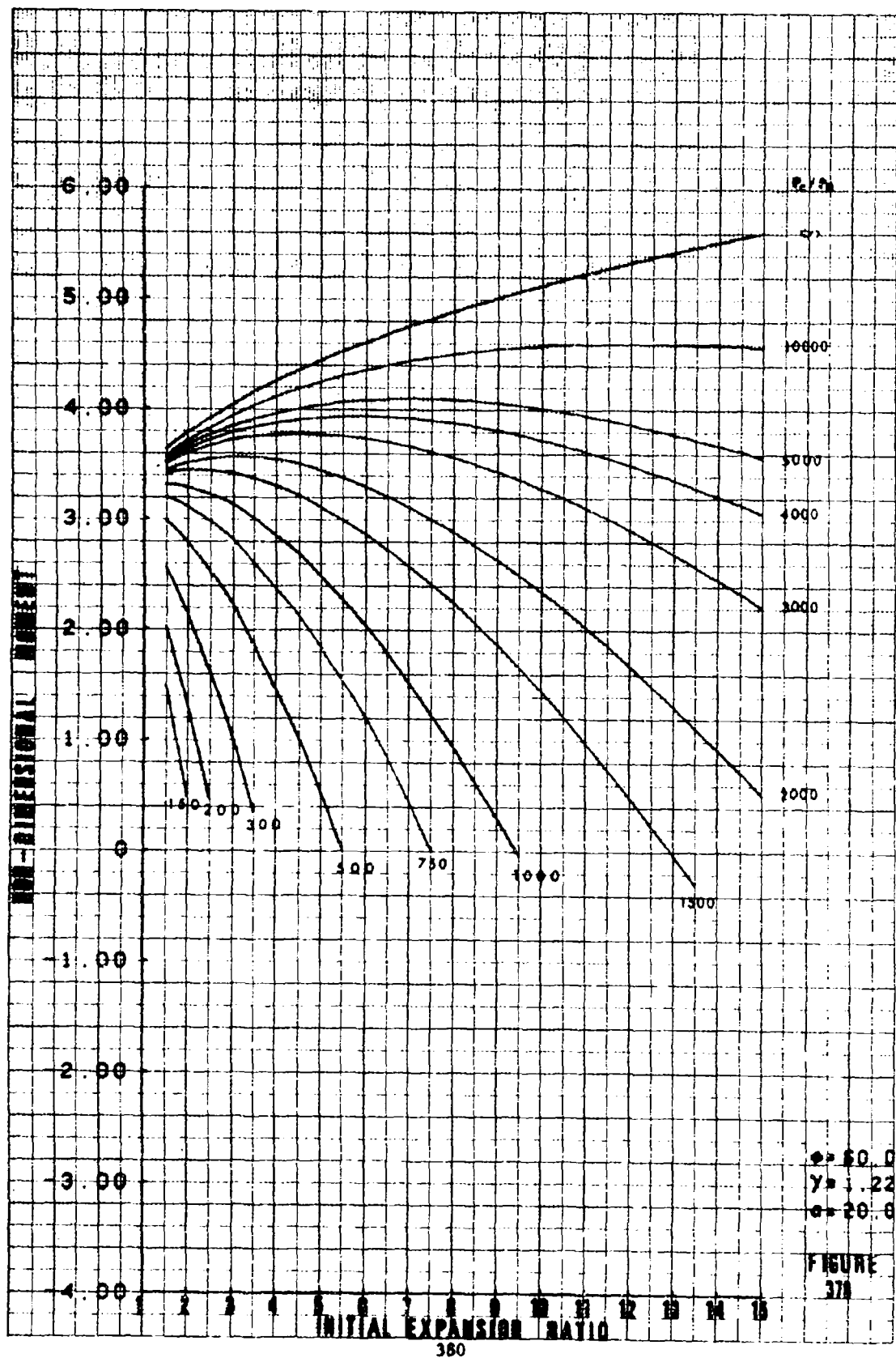


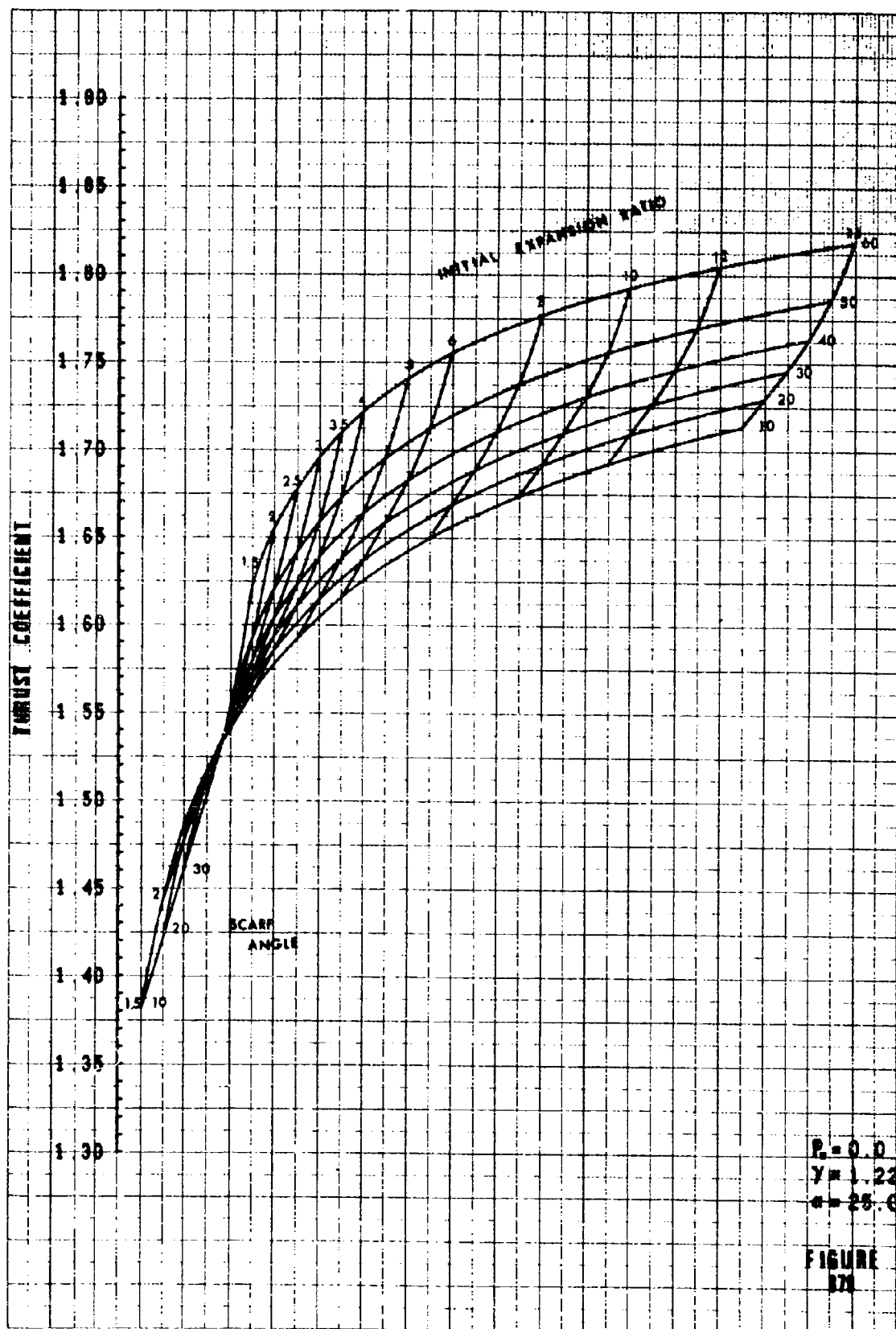
FIGURE 374

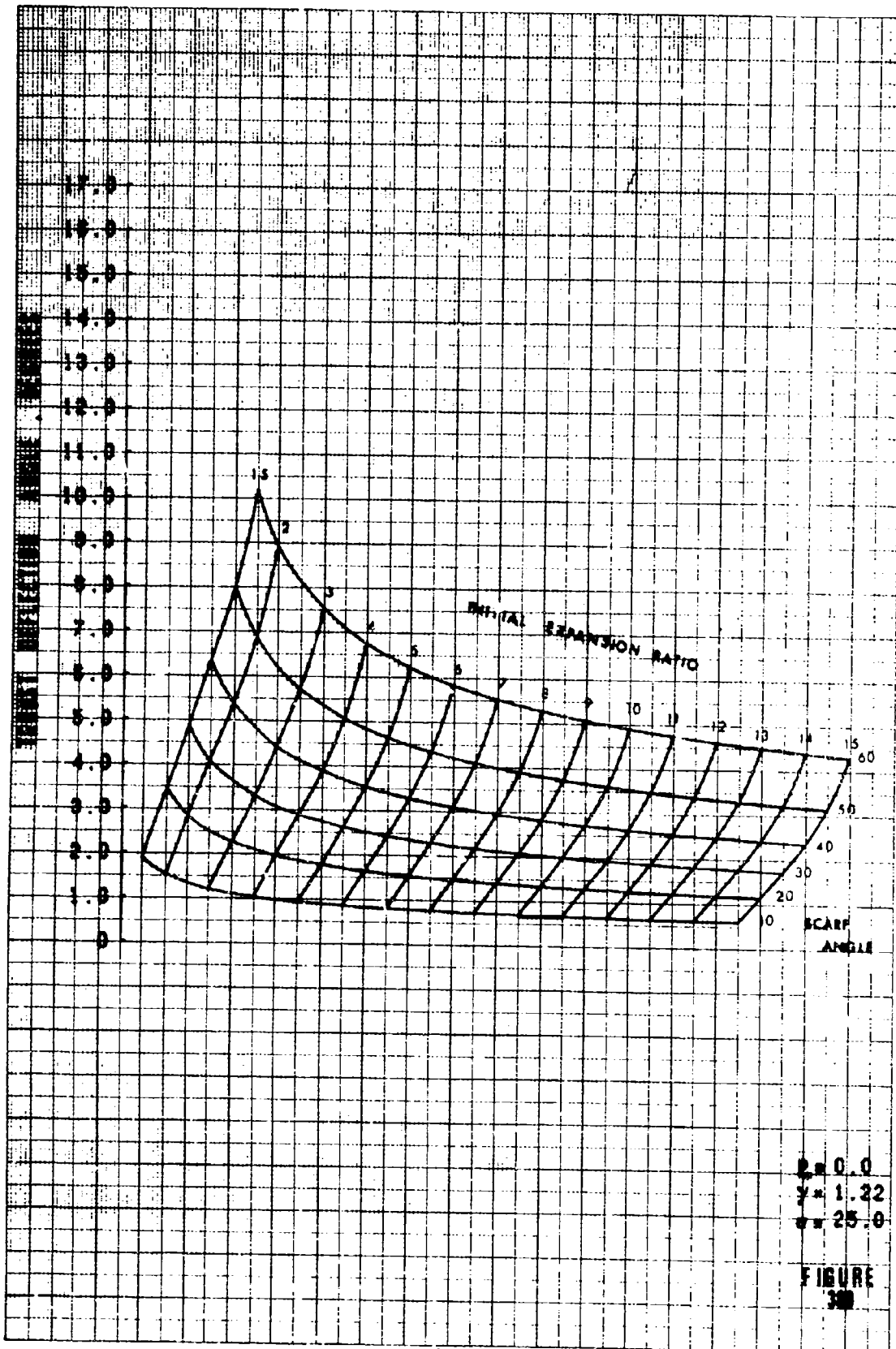


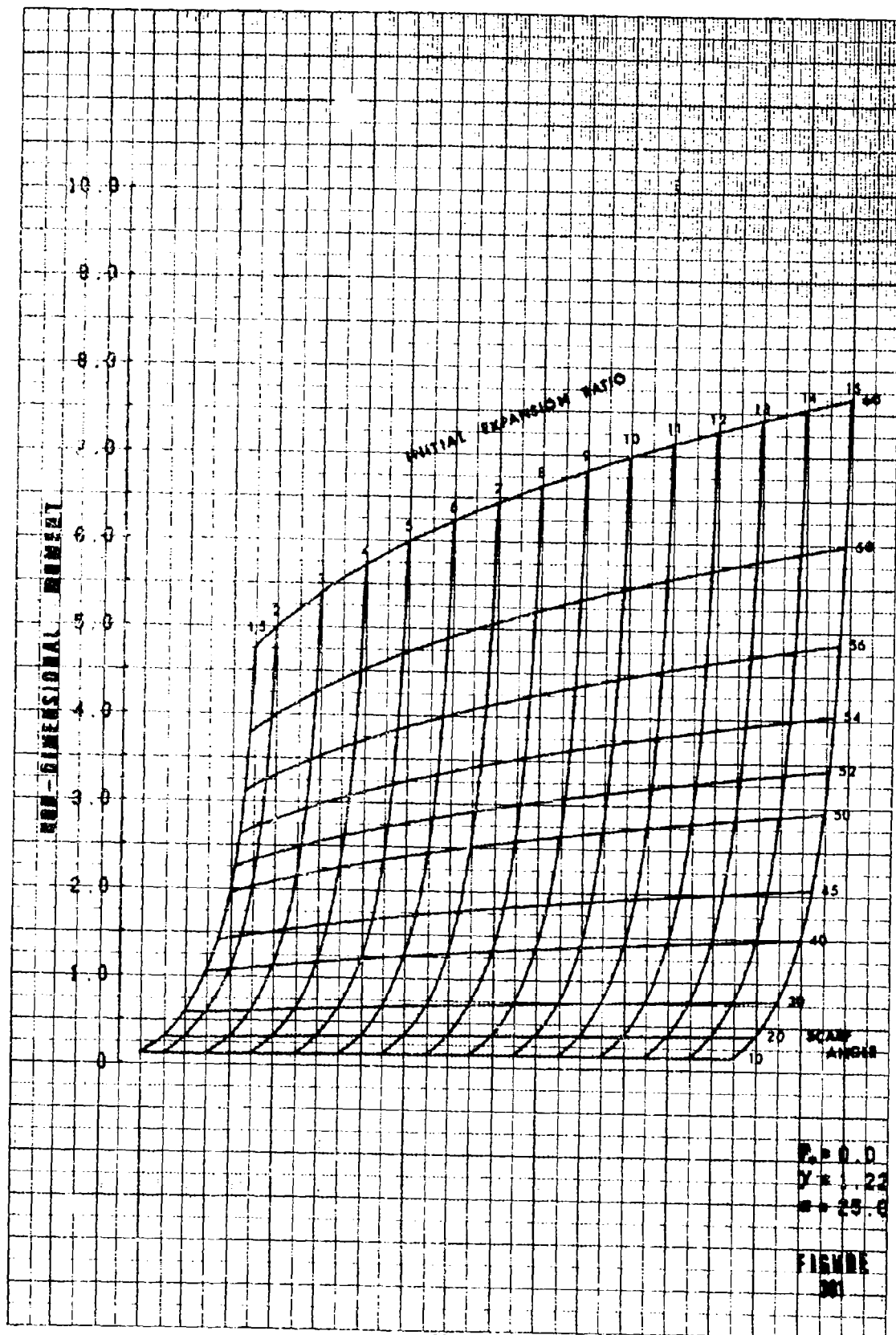


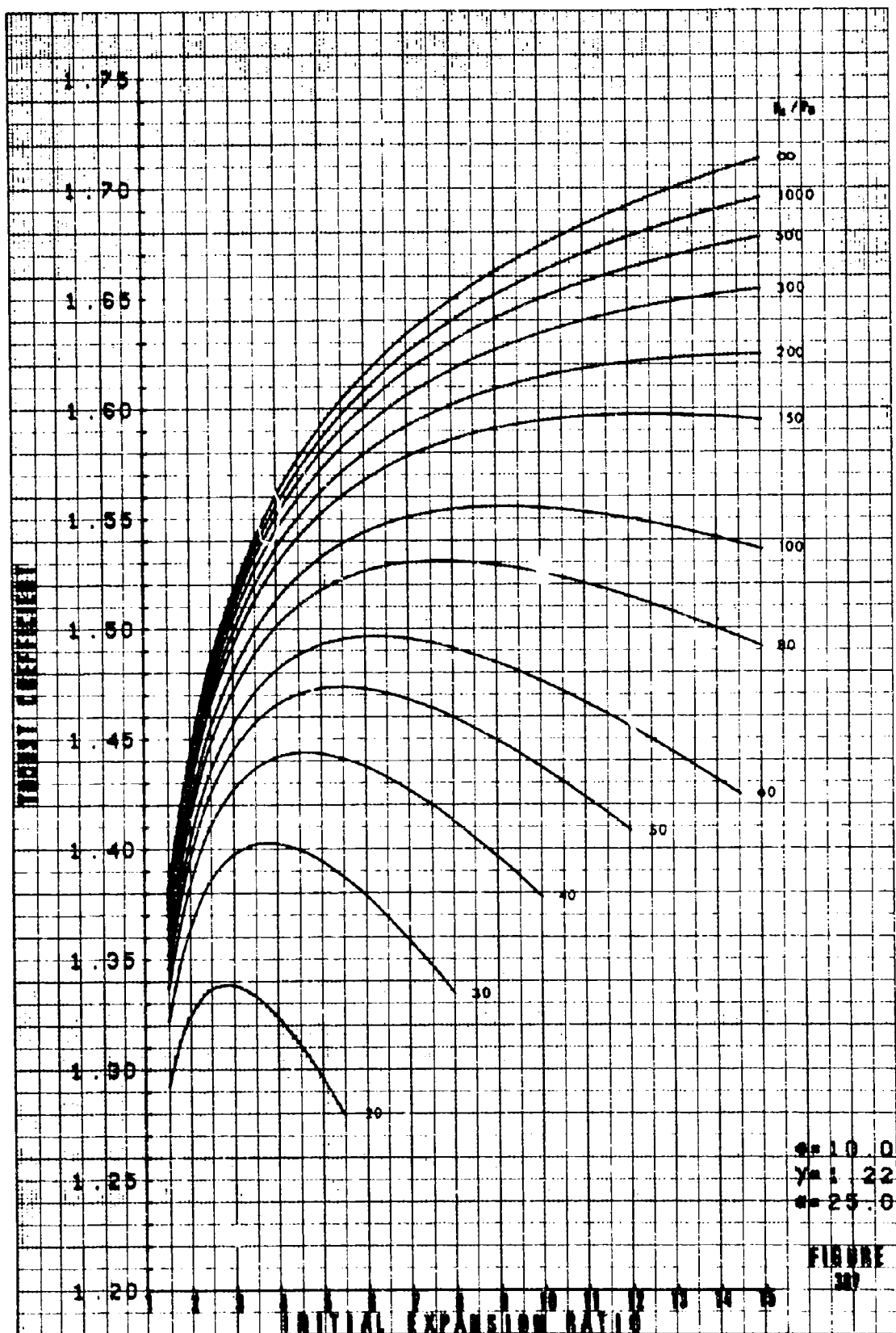






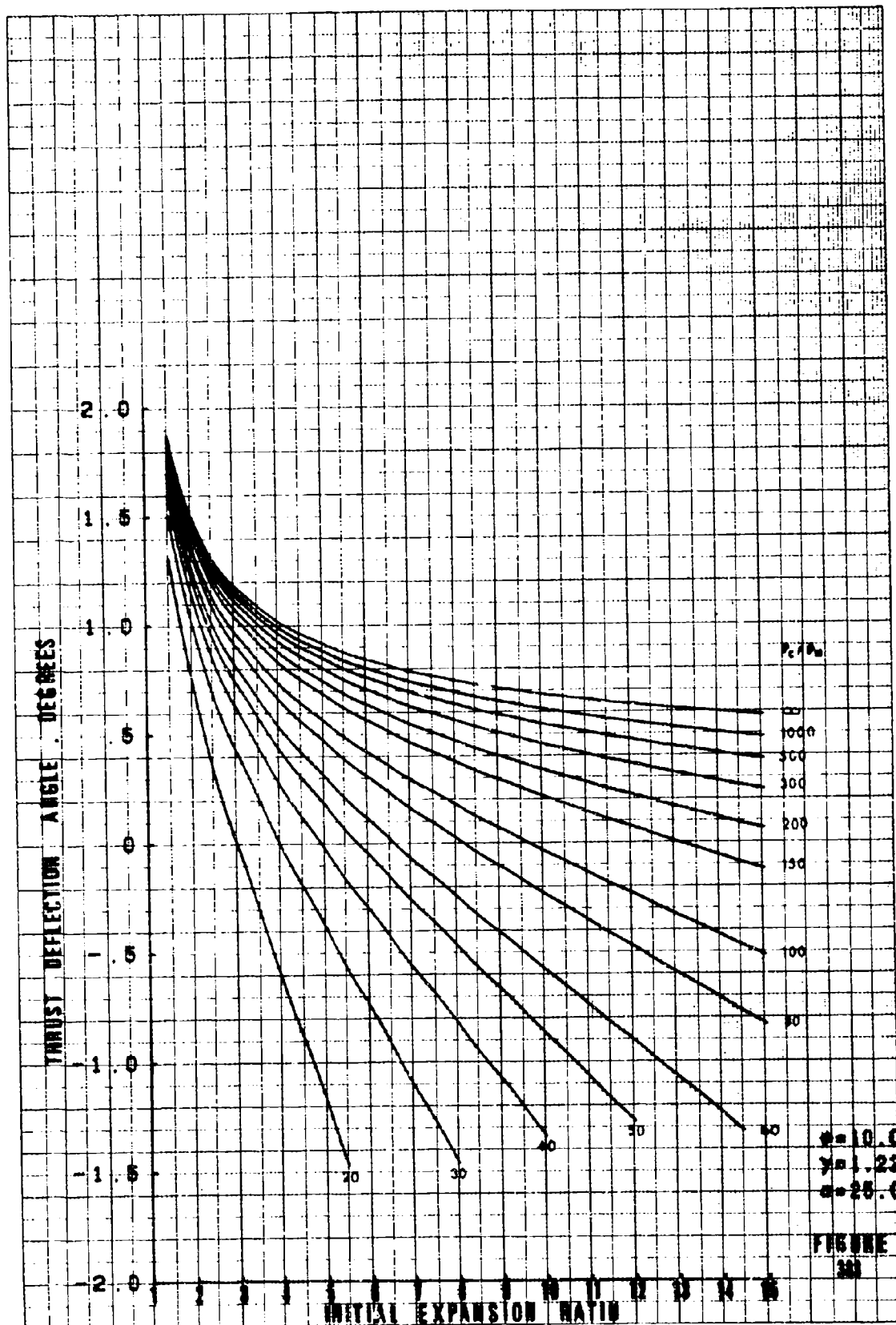






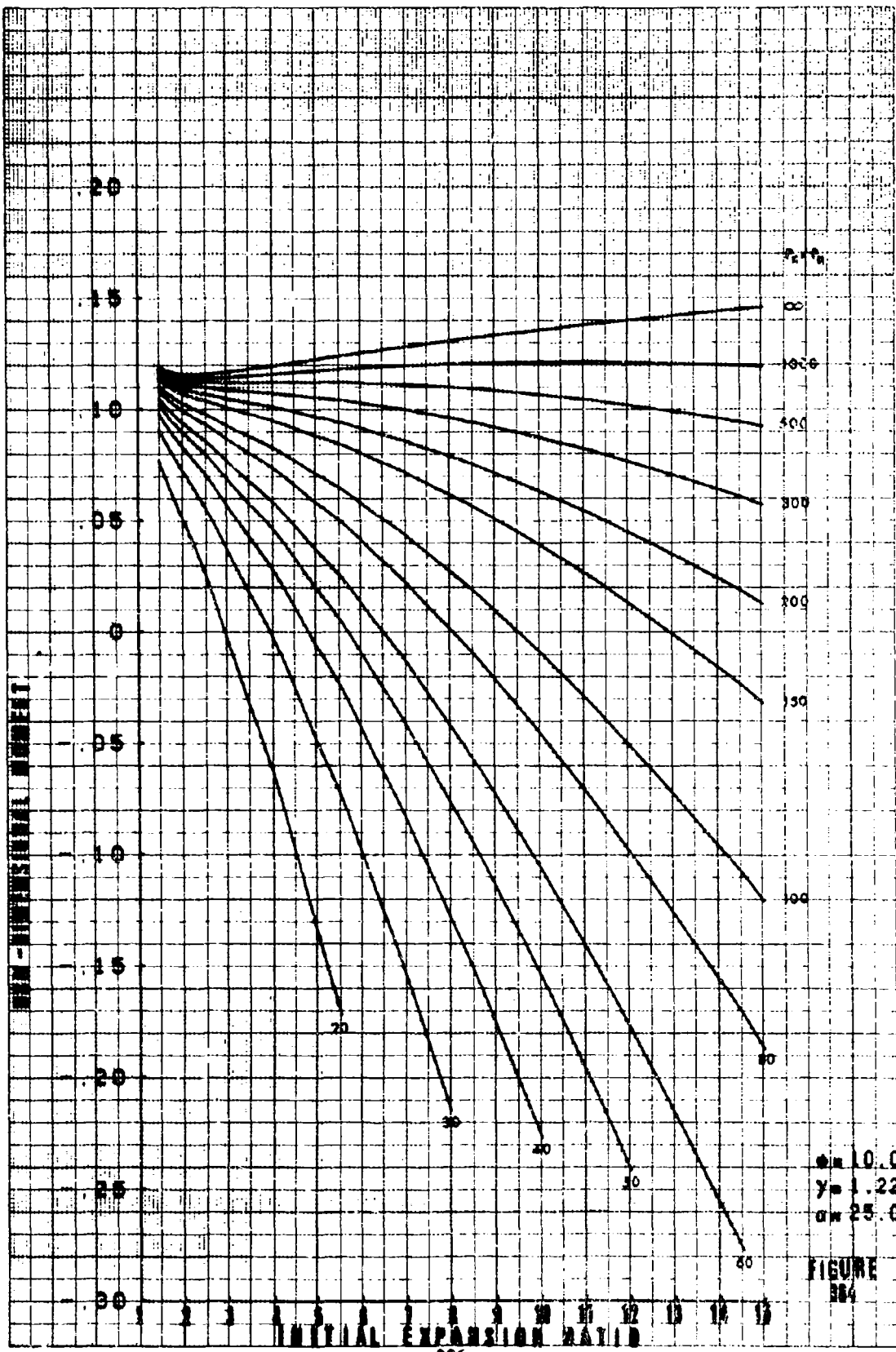
$\gamma = 1.0$
 $\gamma = 1.22$
 $\gamma = 25.0$

FIGURE 307



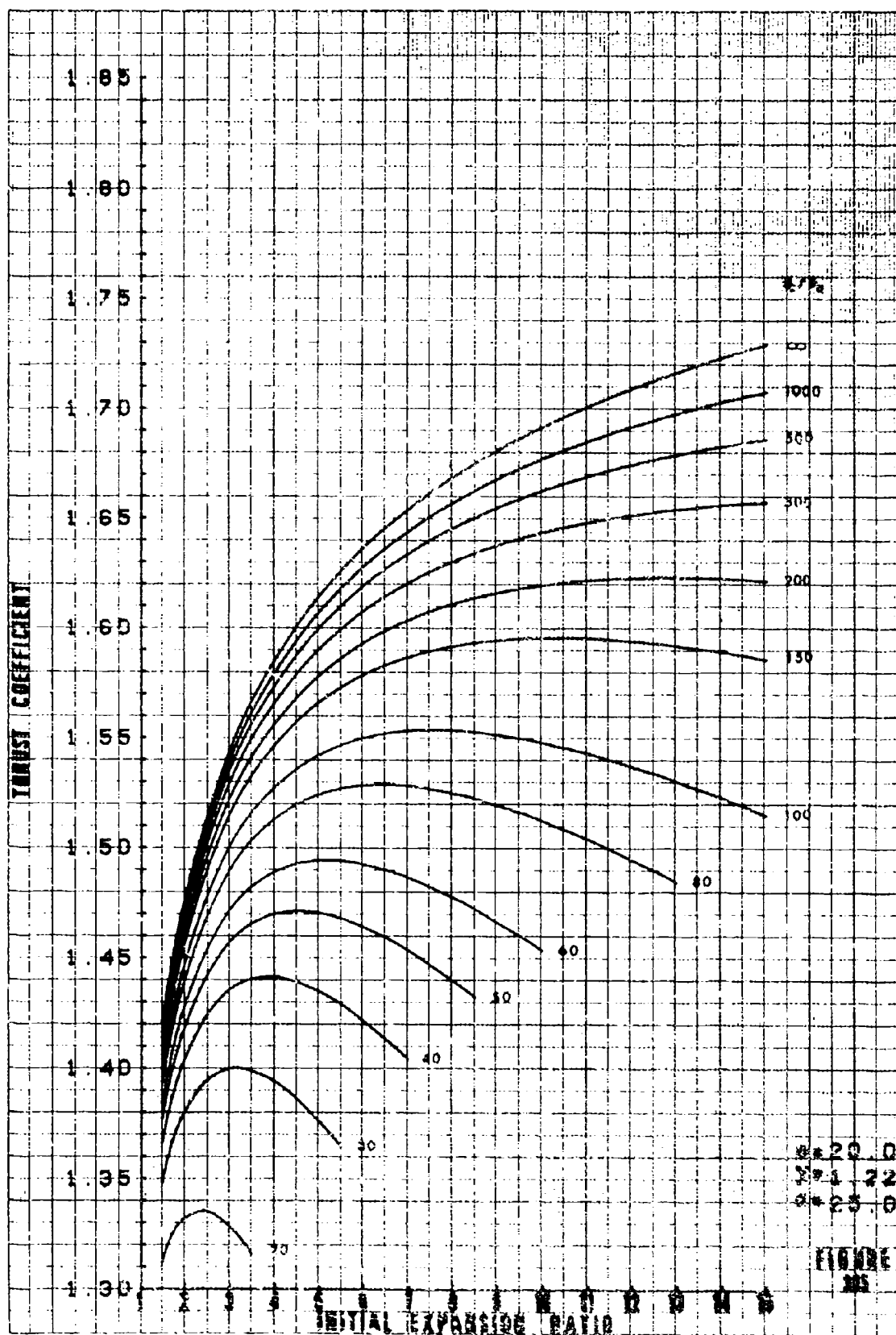
$\gamma = 10.0$
 $\gamma = 1.22$
 $\alpha = 25.0$

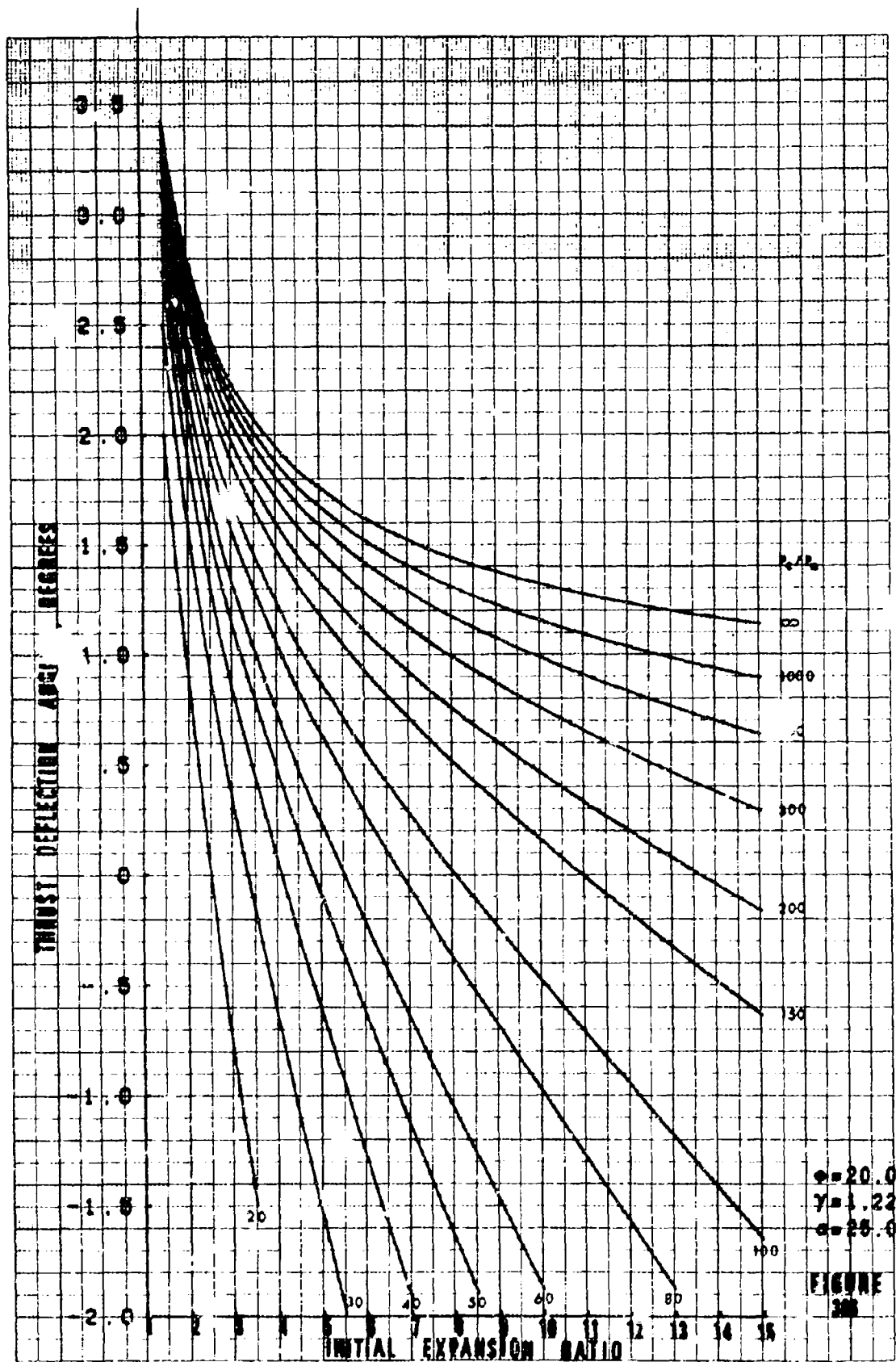
FIGURE 385



$\phi = 10.0$
 $\gamma = 1.22$
 $\sigma = 25.0$

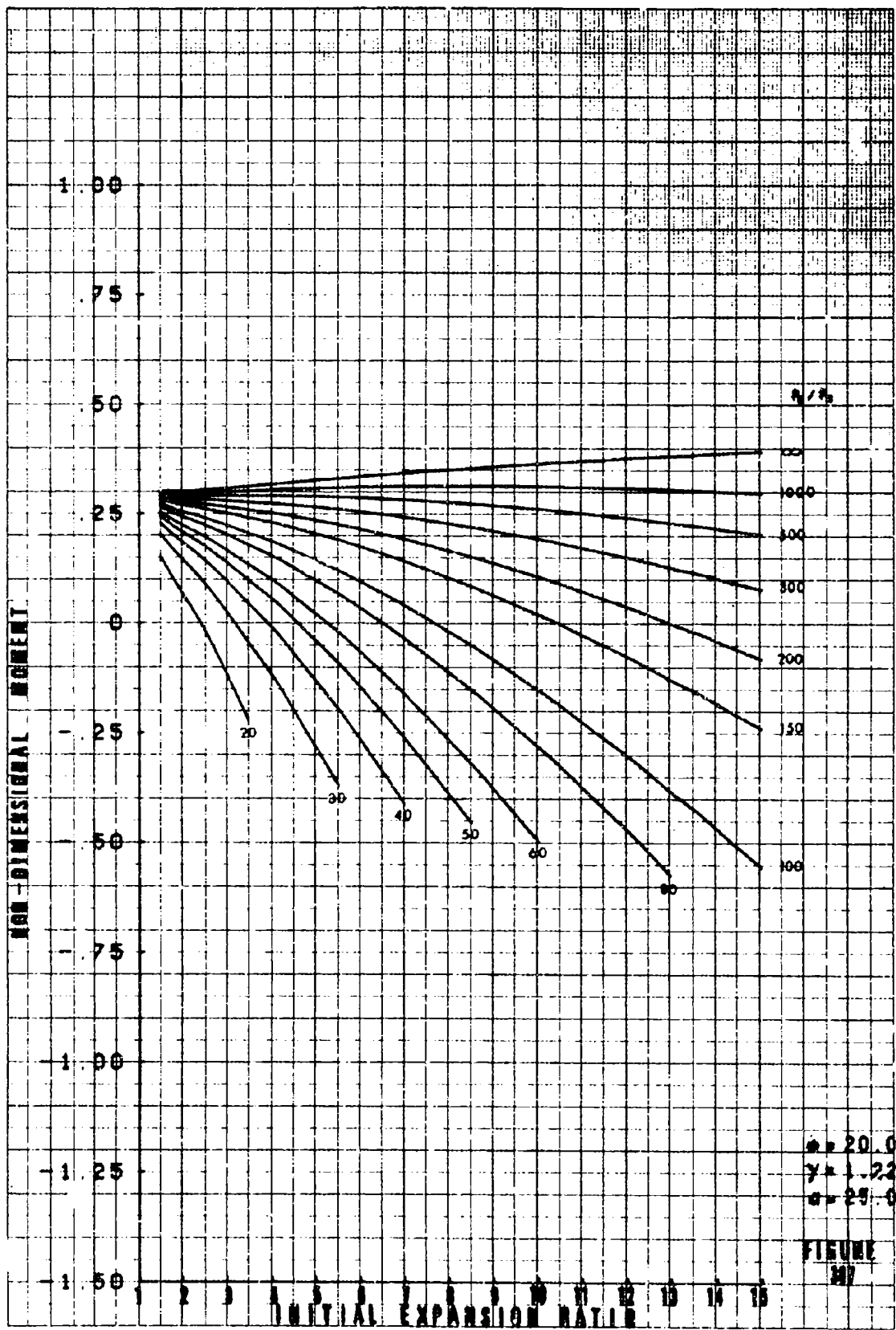
FIGURE 386





$\gamma = 20.0$
 $\gamma = 1.22$
 $\gamma = 25.0$

FIGURE 200



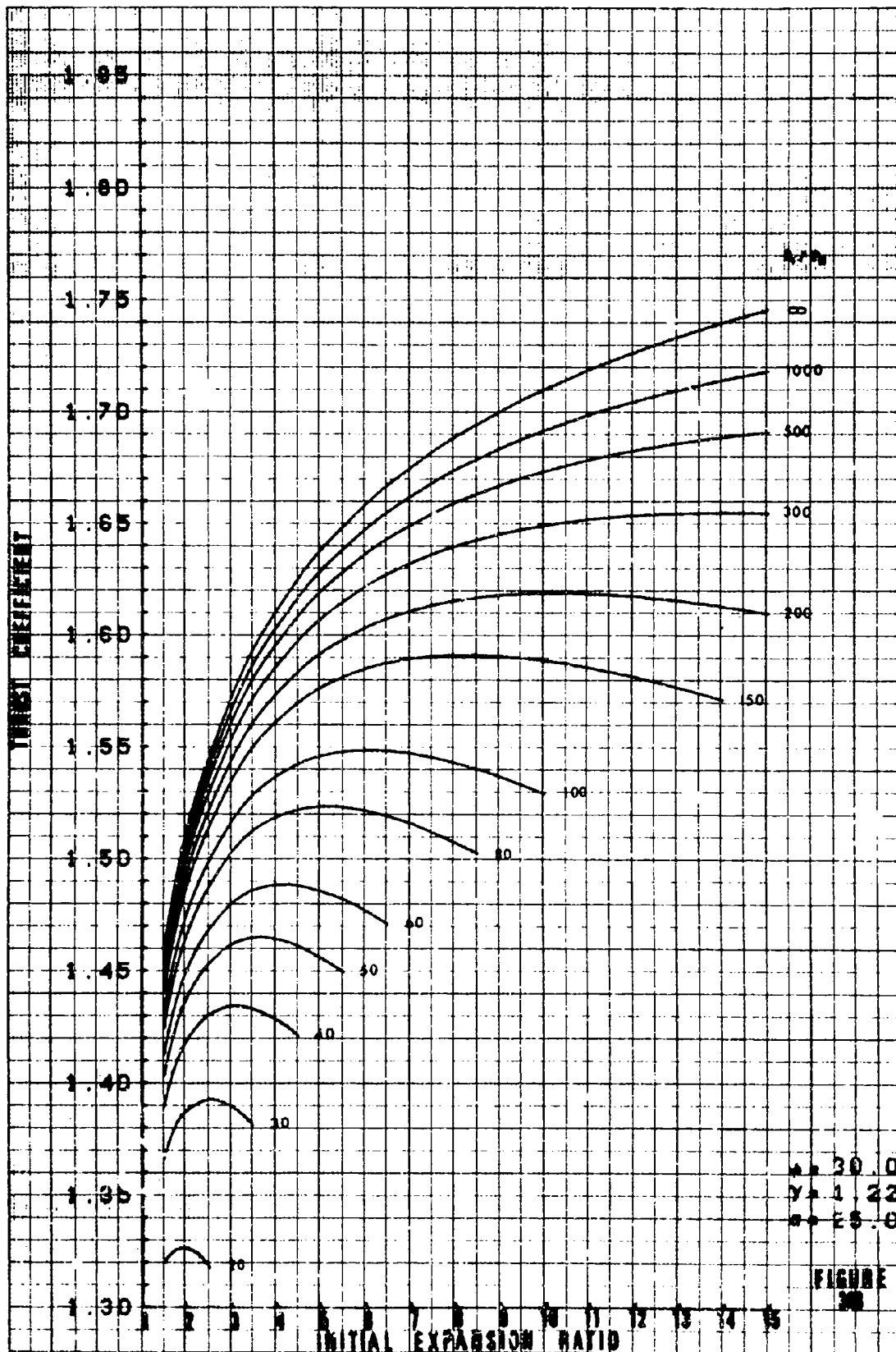
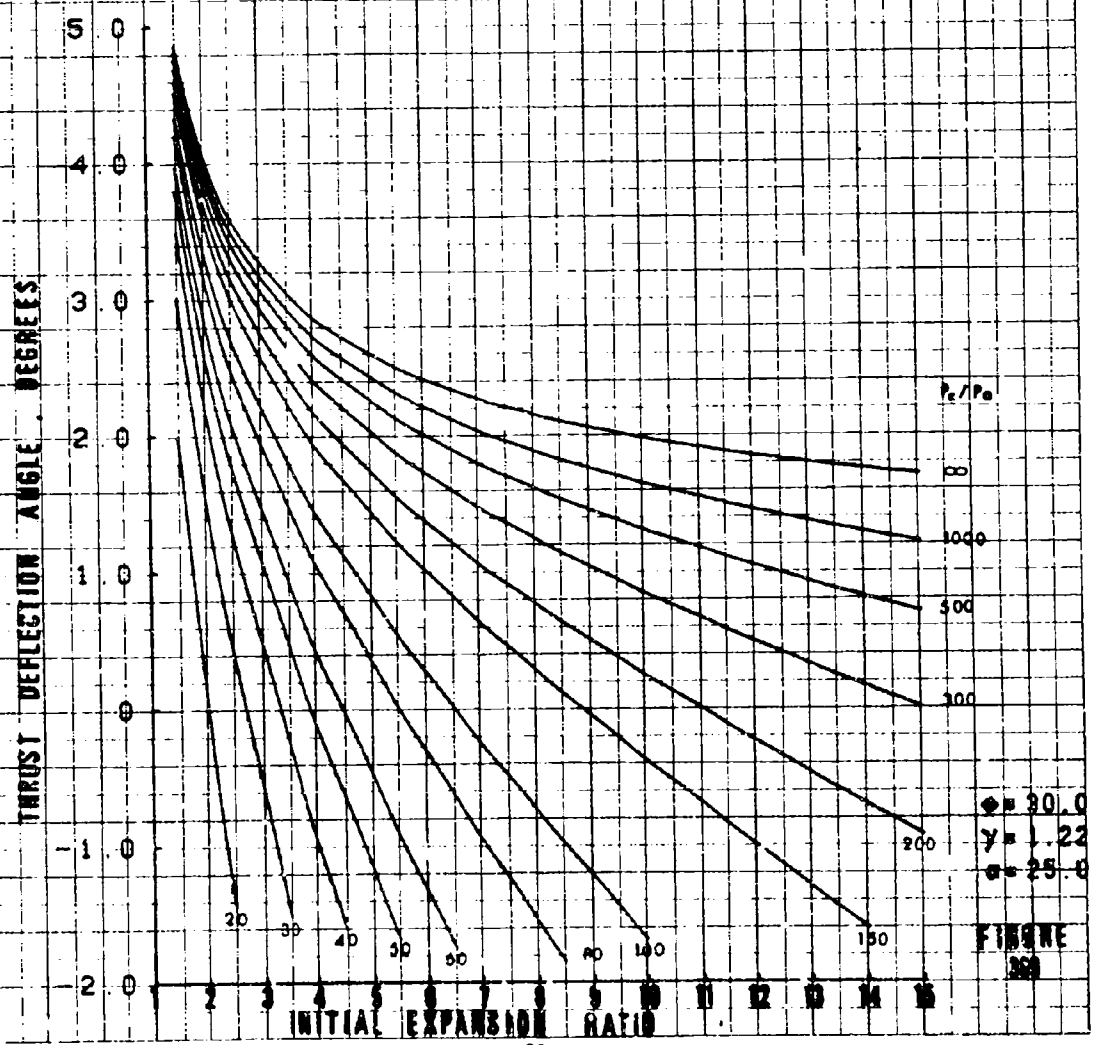
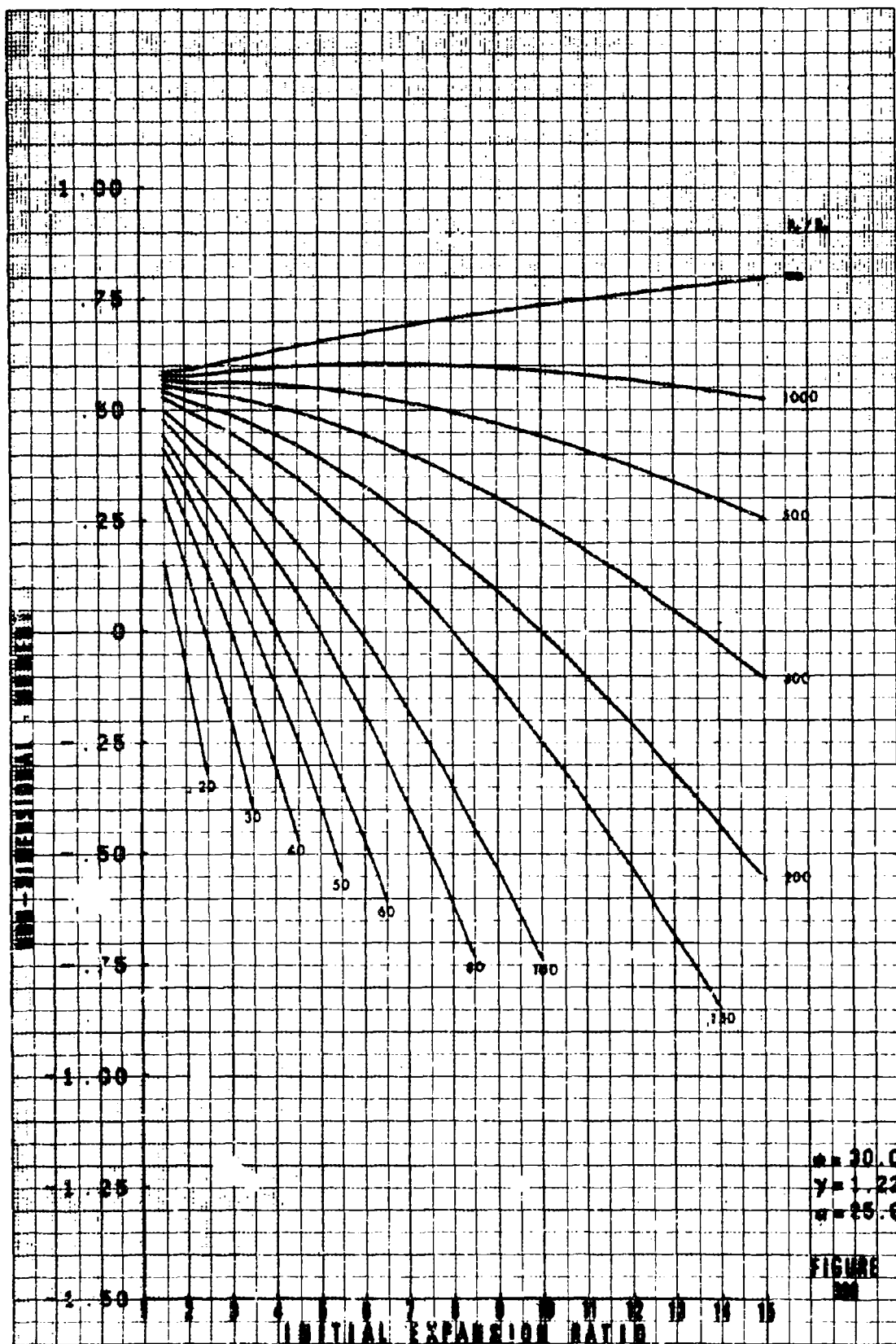


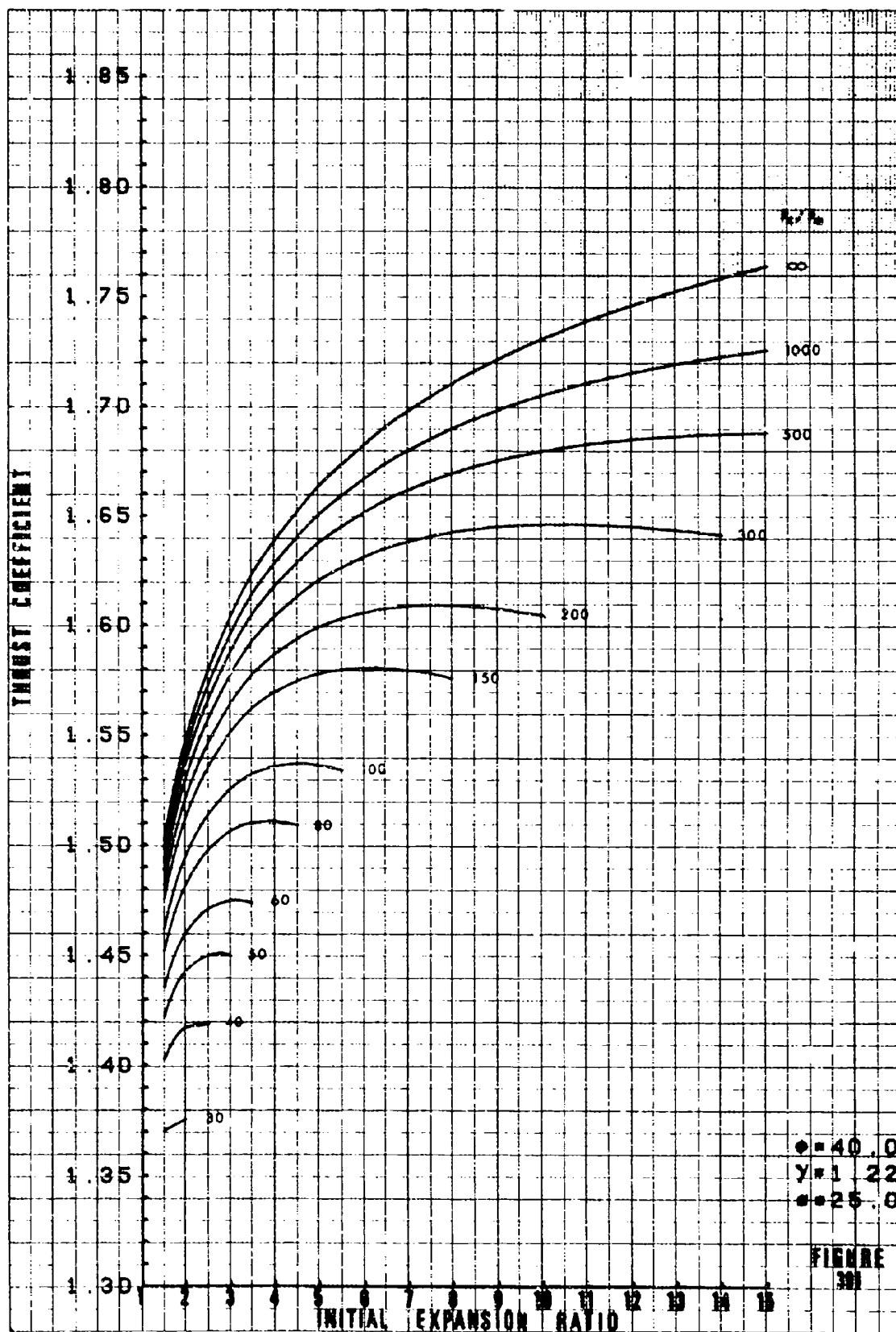
FIGURE 20

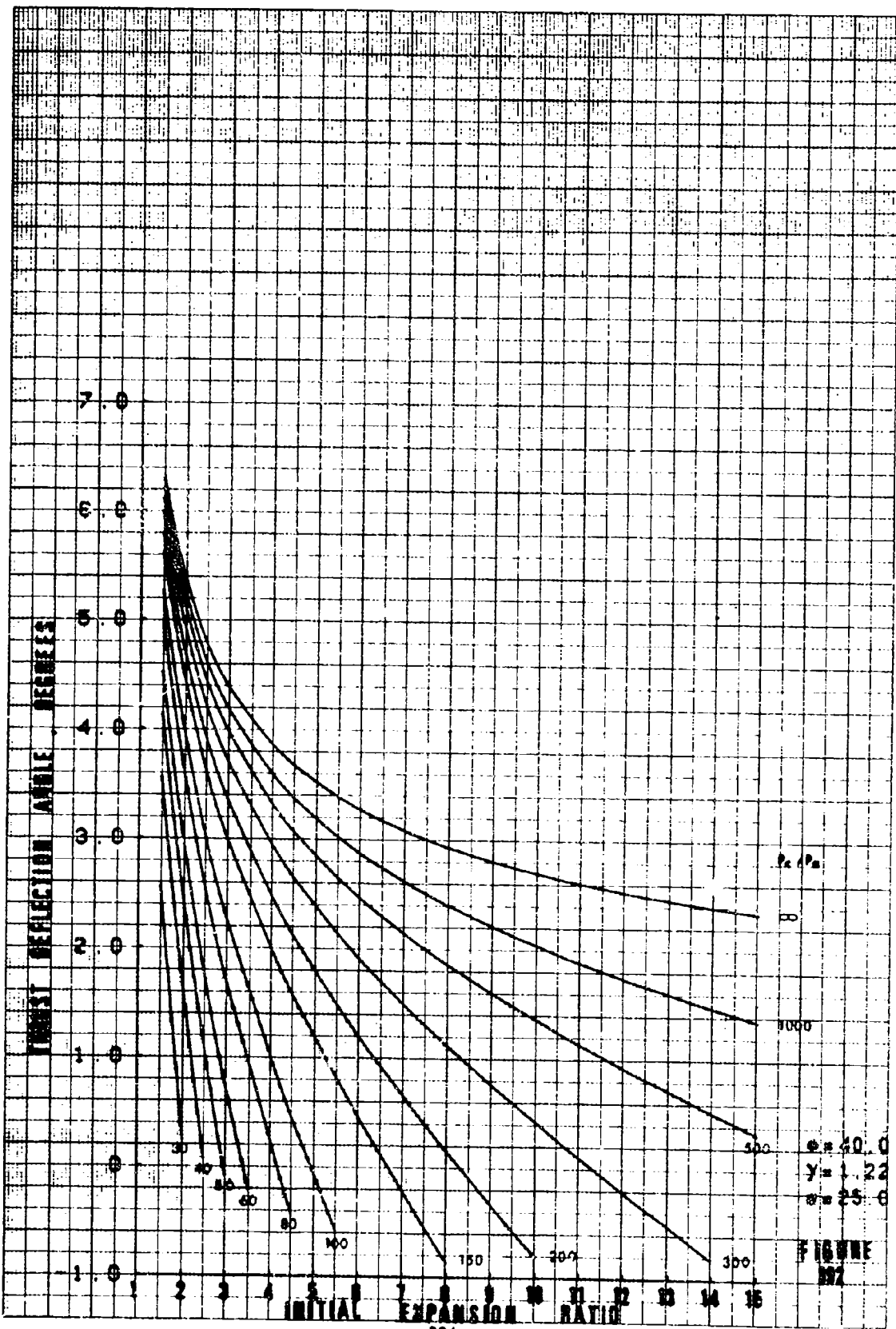


$\phi = 30.0$
 $\gamma = 1.22$
 $\alpha = 25.0$

FIGURE 259

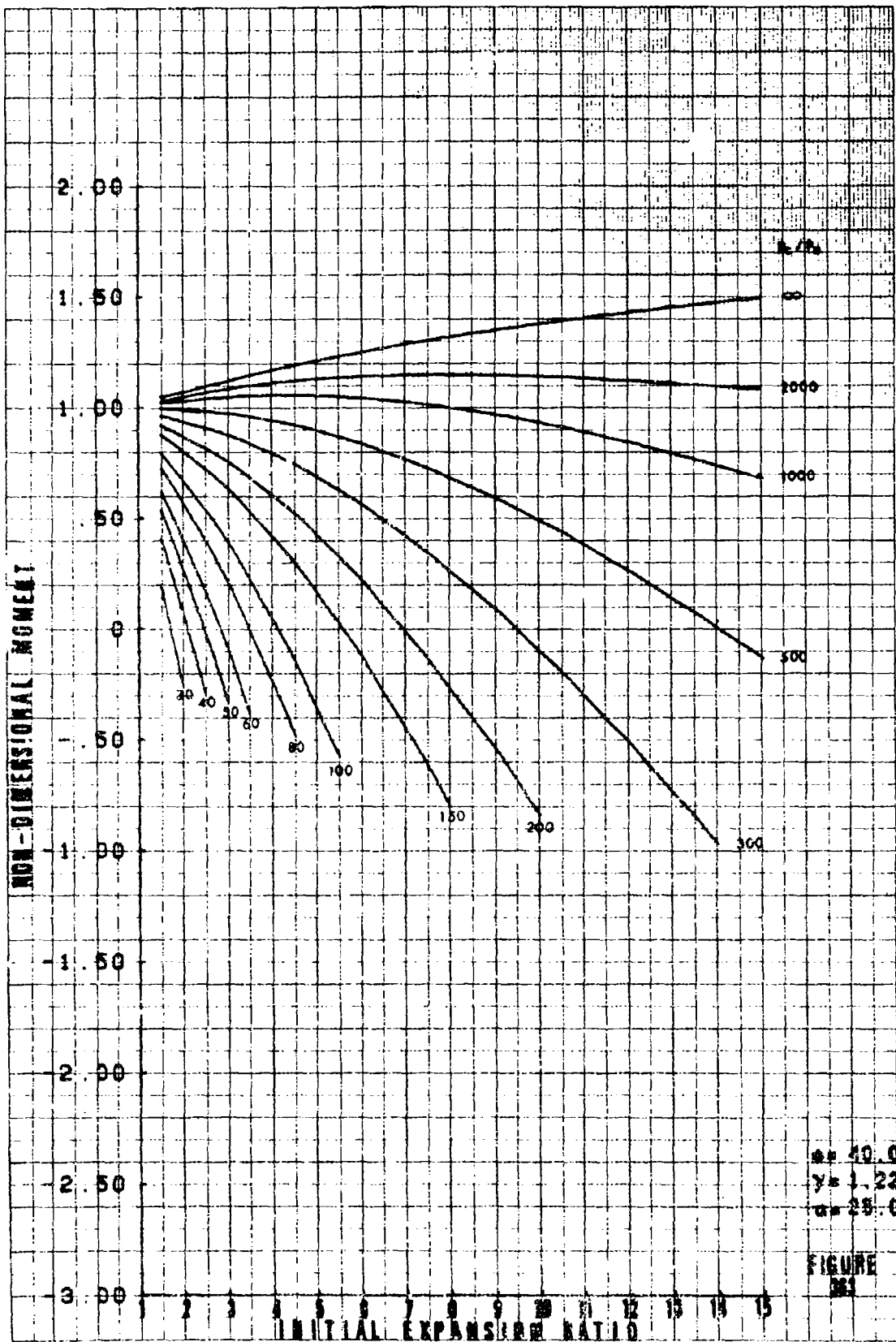






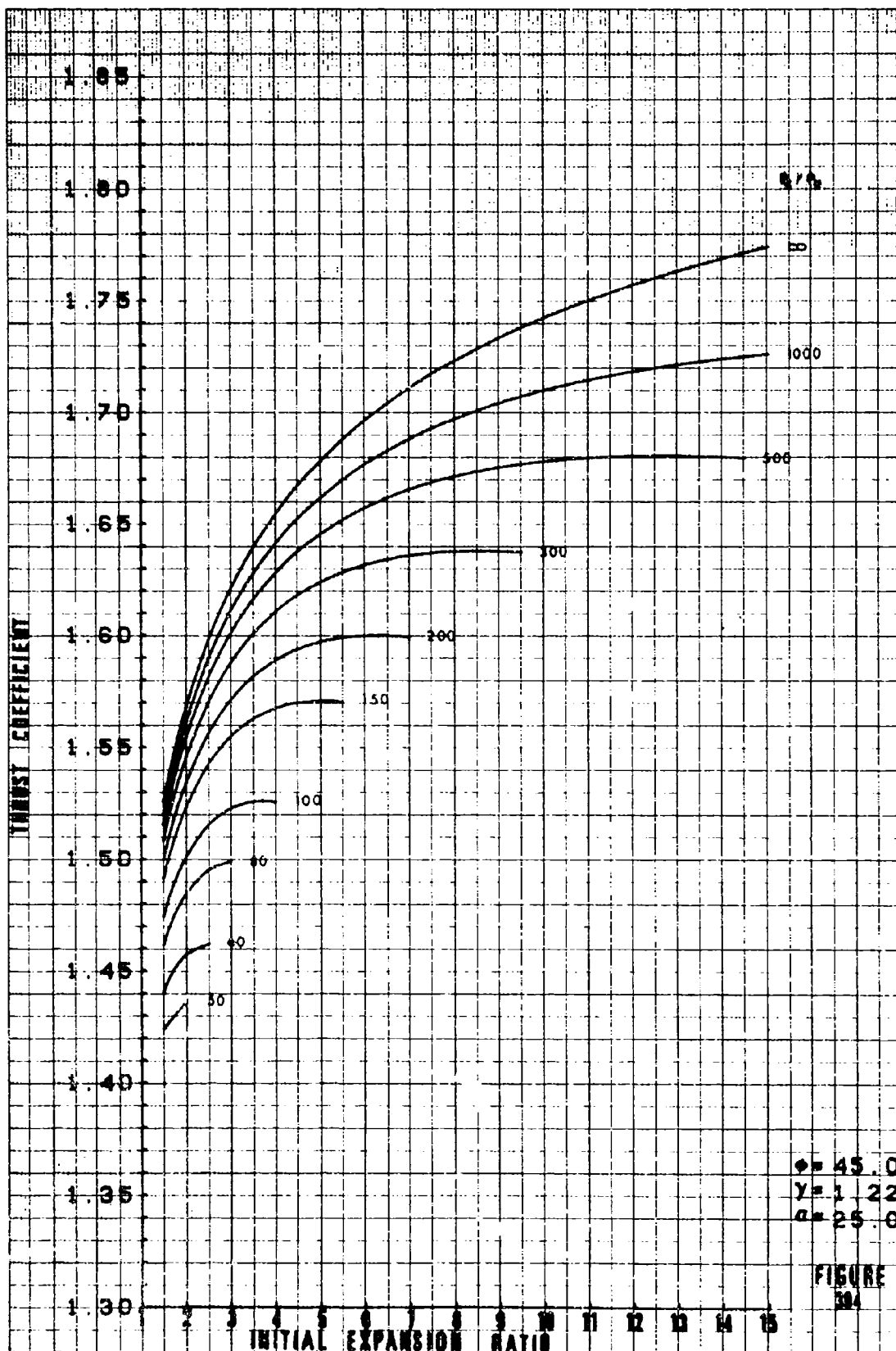
$\phi = 40.0$
 $\gamma = 1.22$
 $\theta = 25.0$

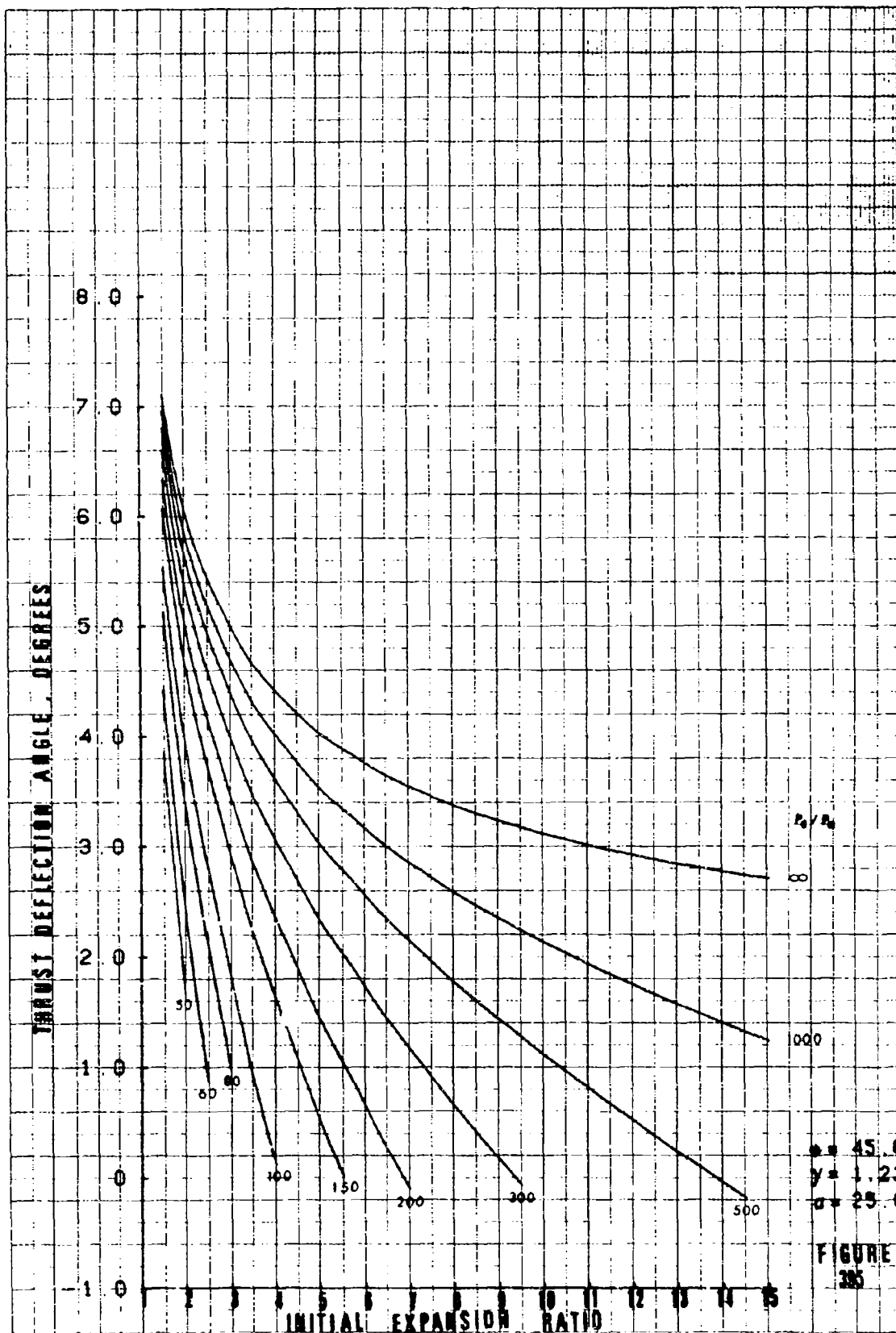
FIGURE 892



$\mu = 40.0$
 $\gamma = 1.22$
 $\alpha = 25.0$

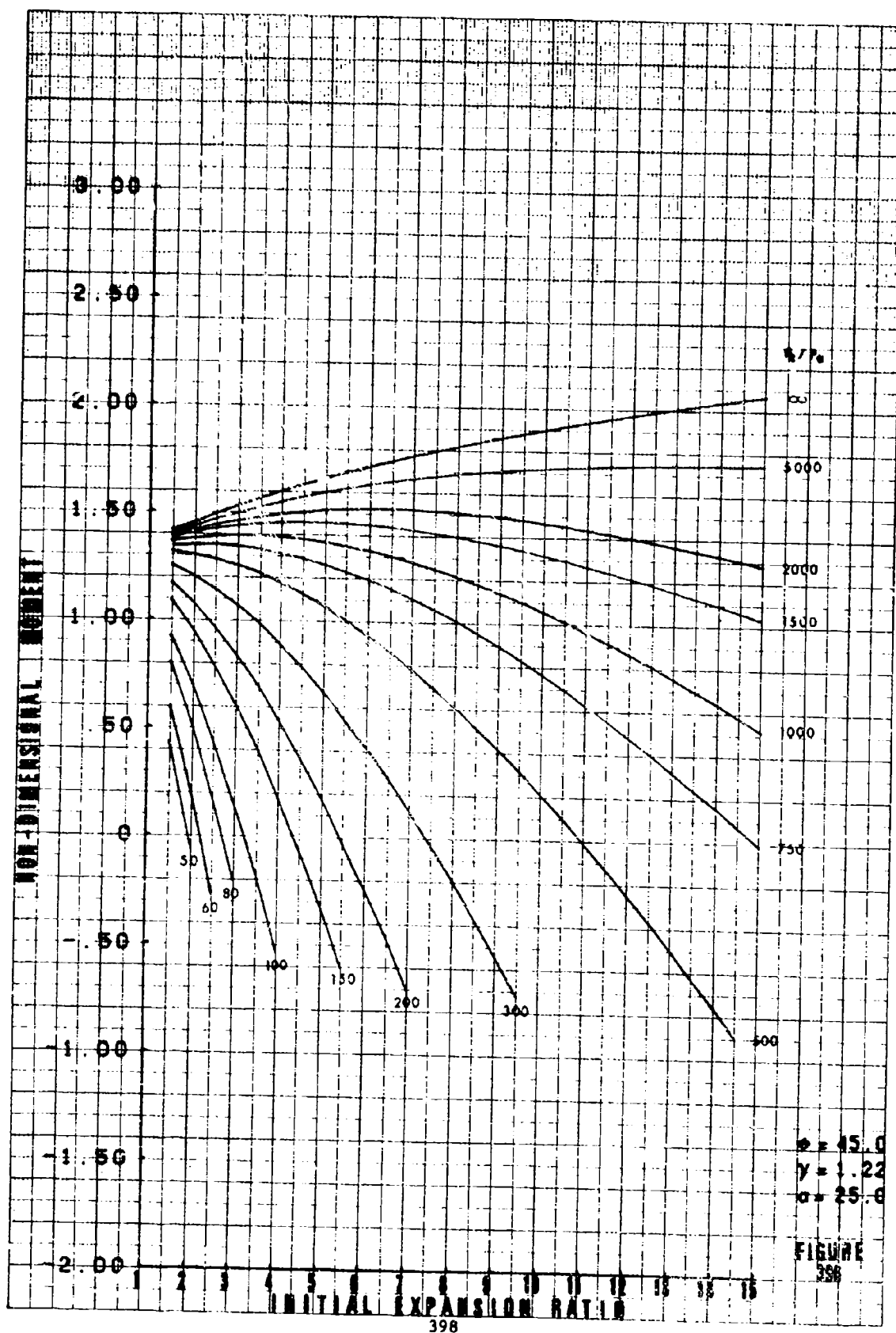
FIGURE 261

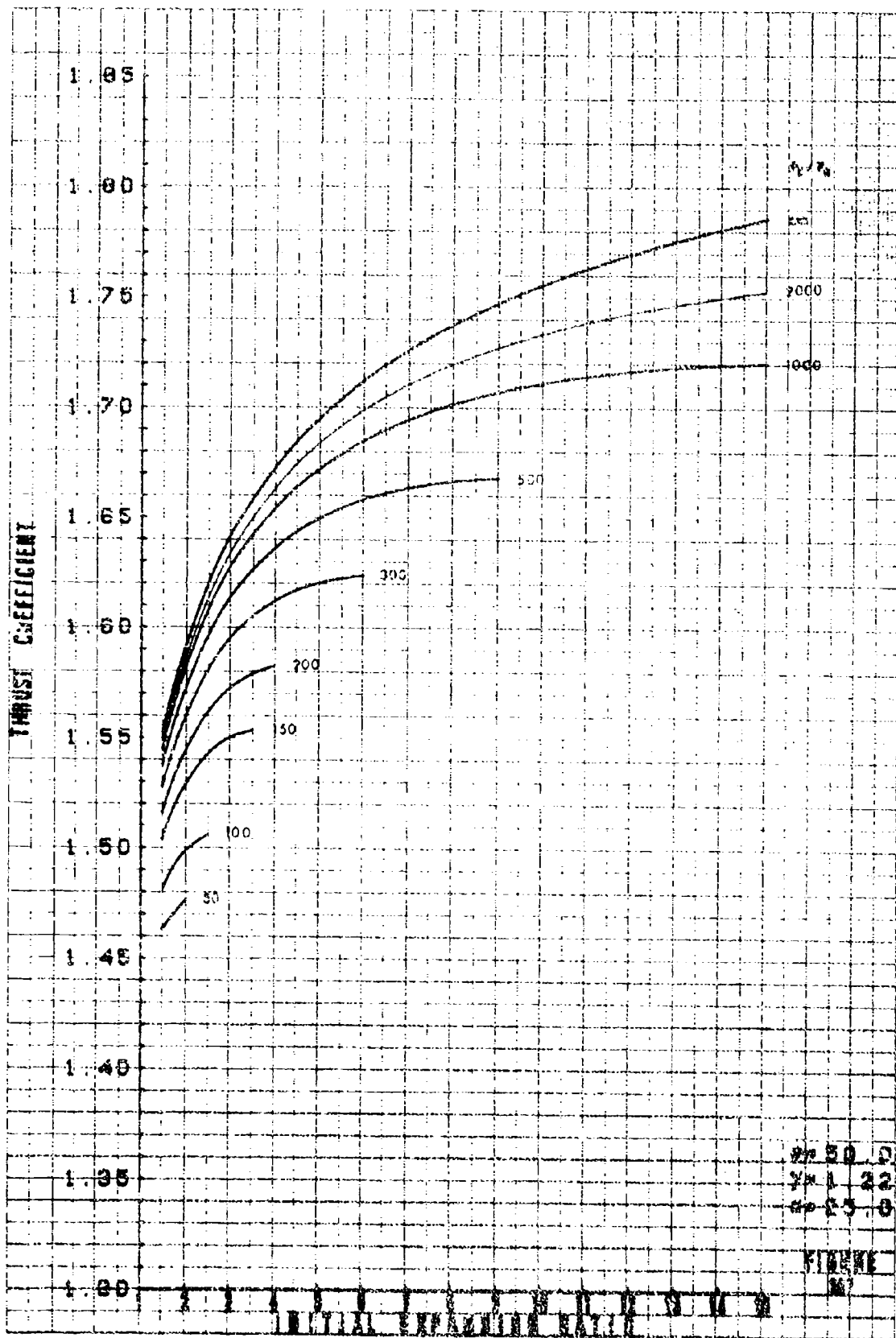


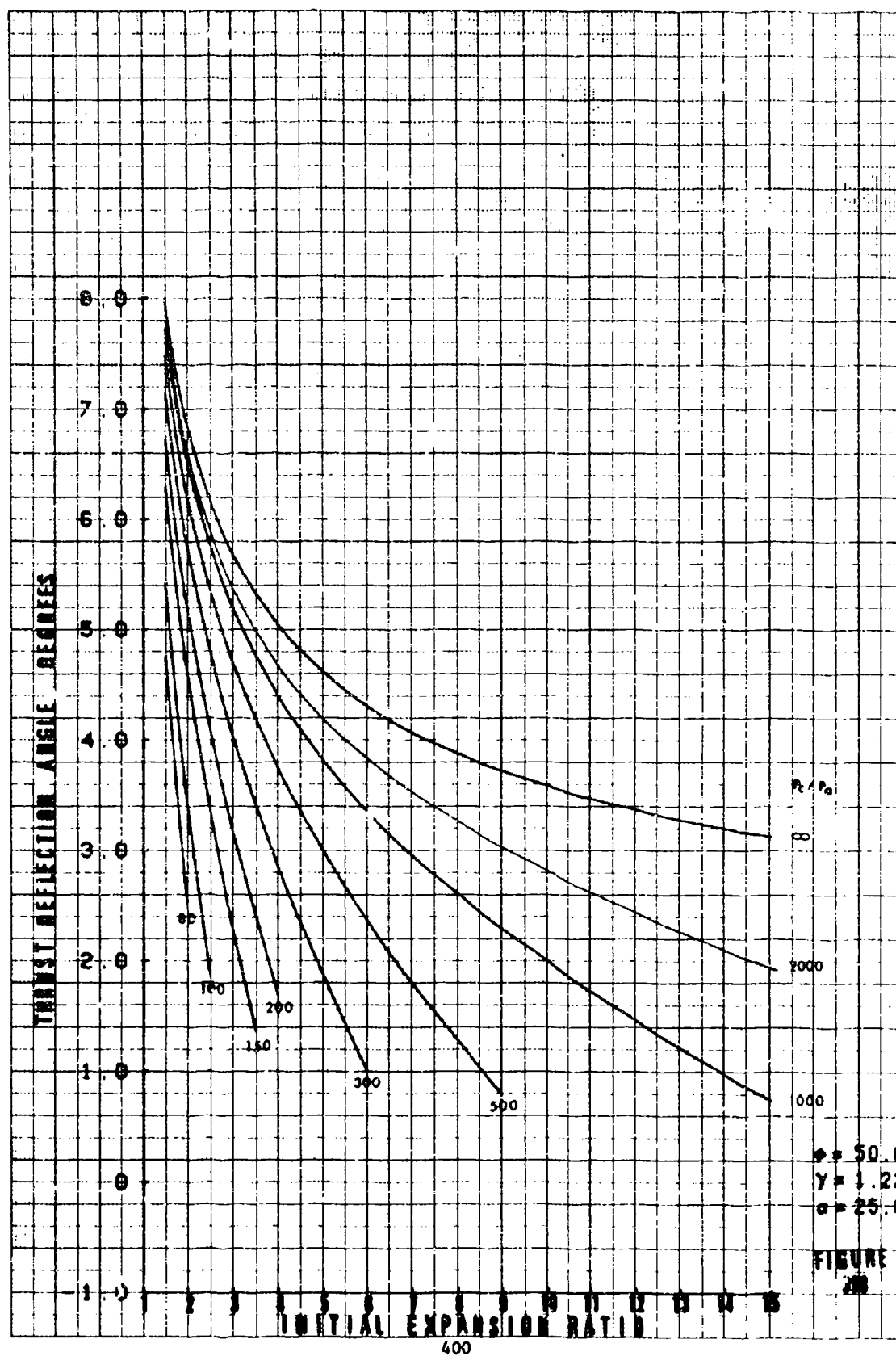


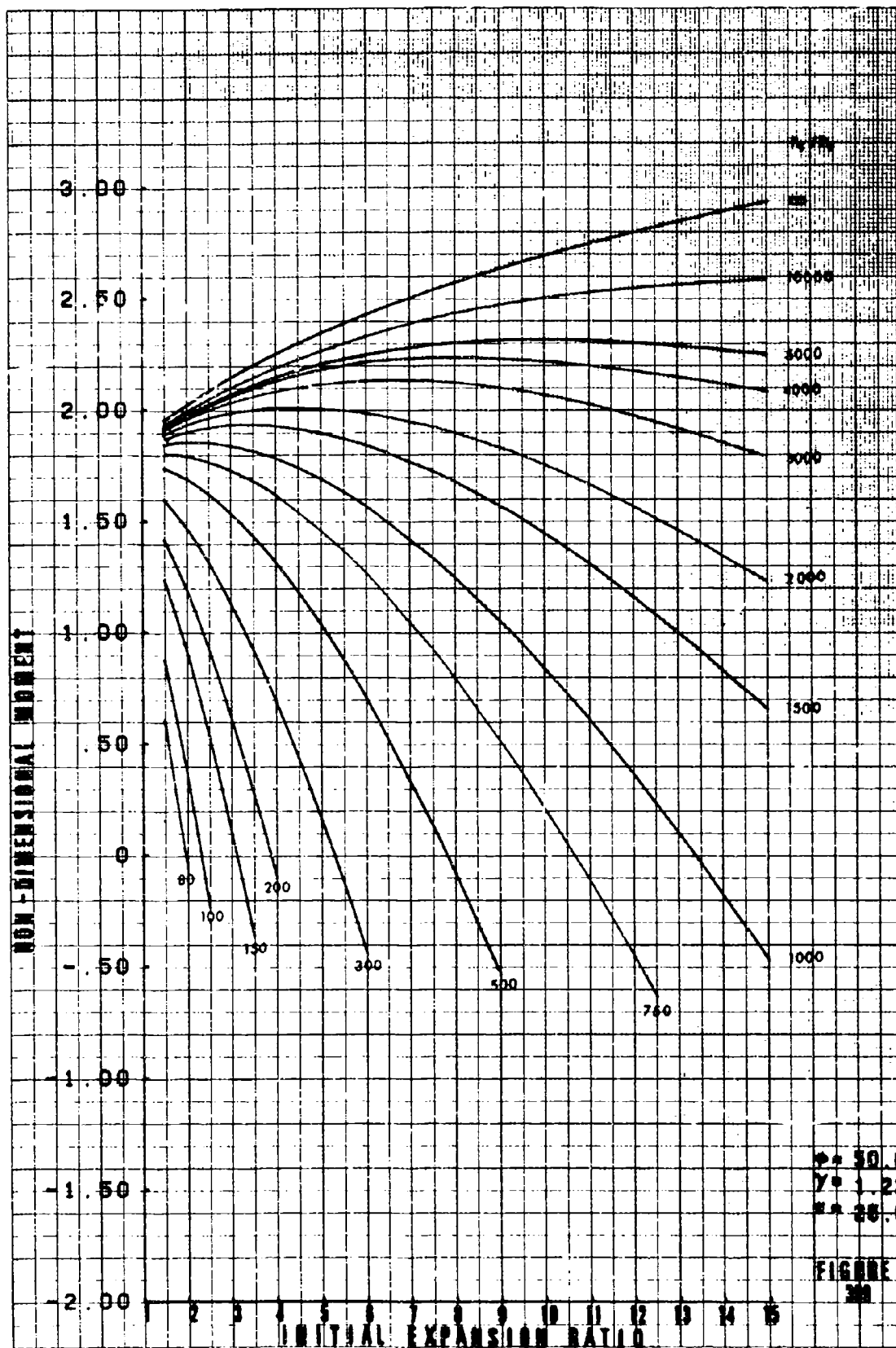
$\gamma = 1.22$
 $\sigma = 25.0$
 $P_0/P_a = 45.0$

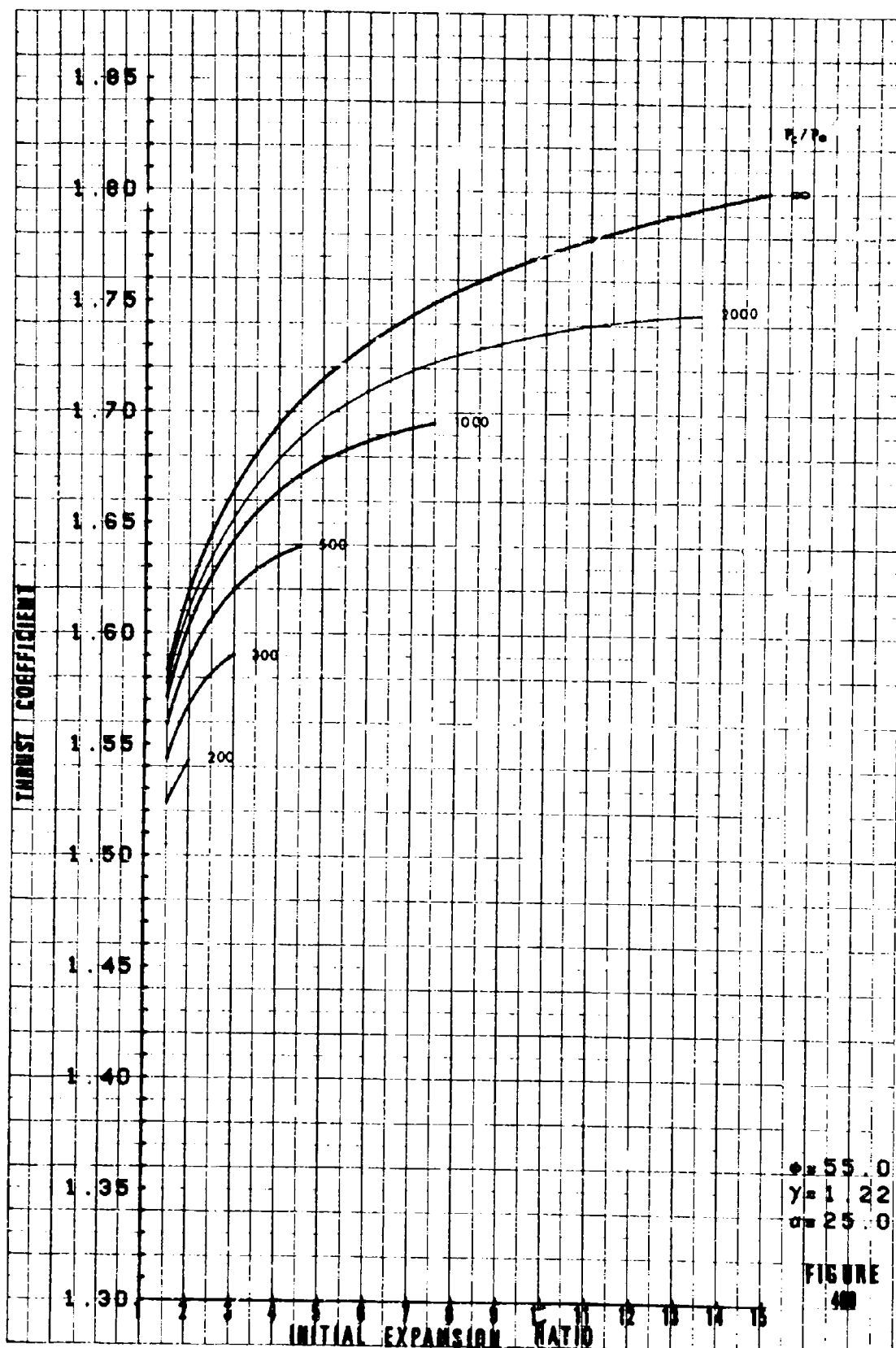
FIGURE 385

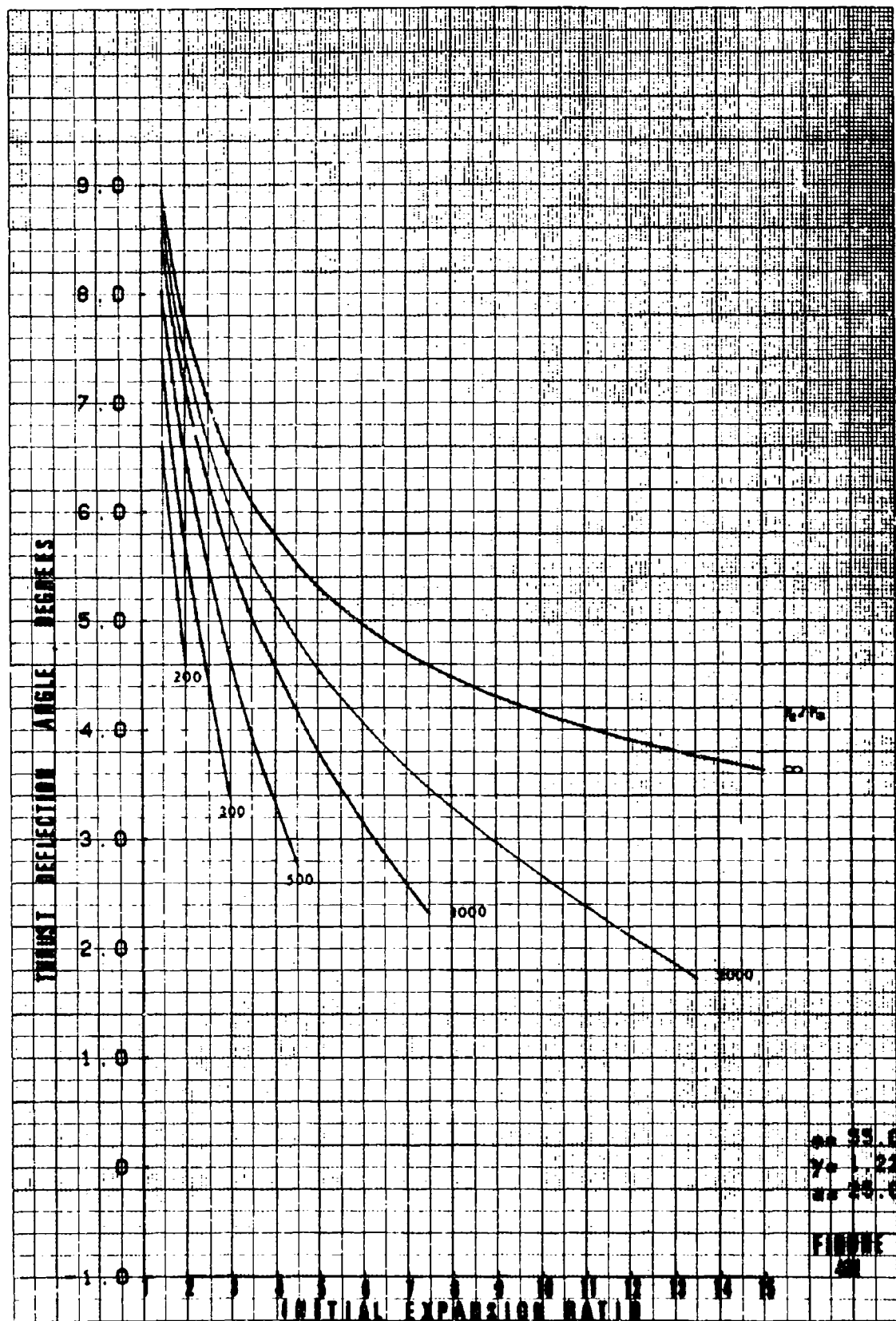


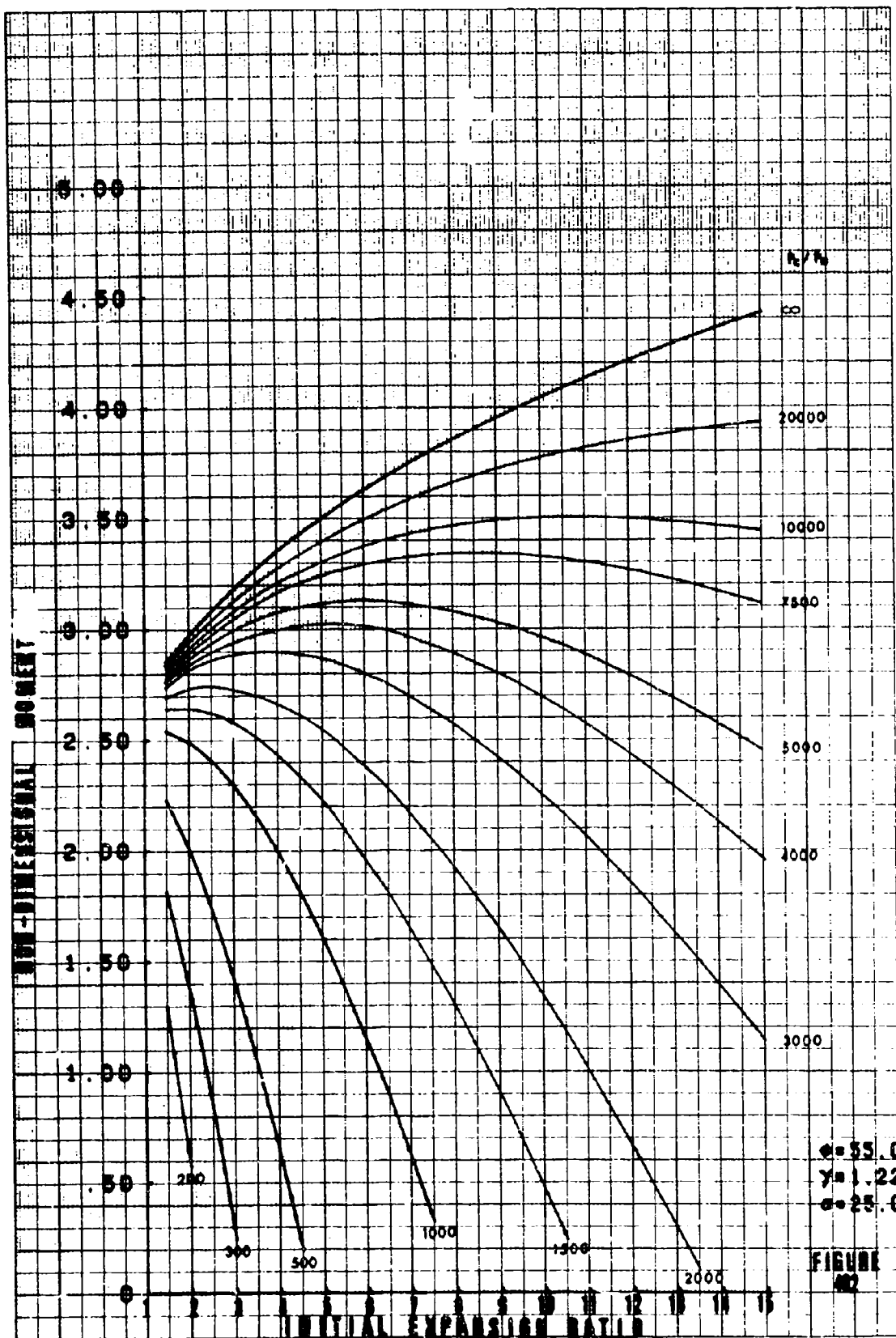


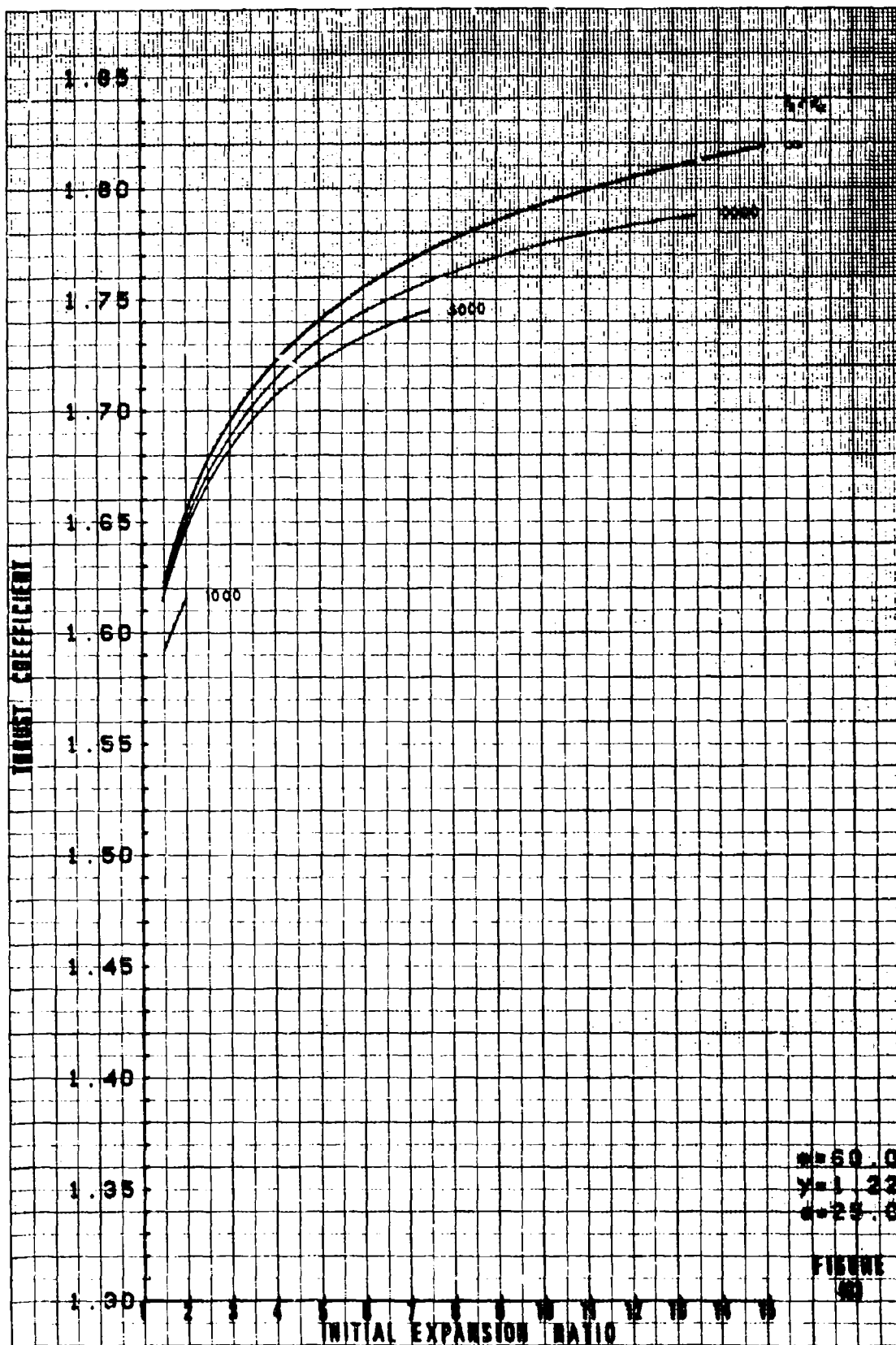


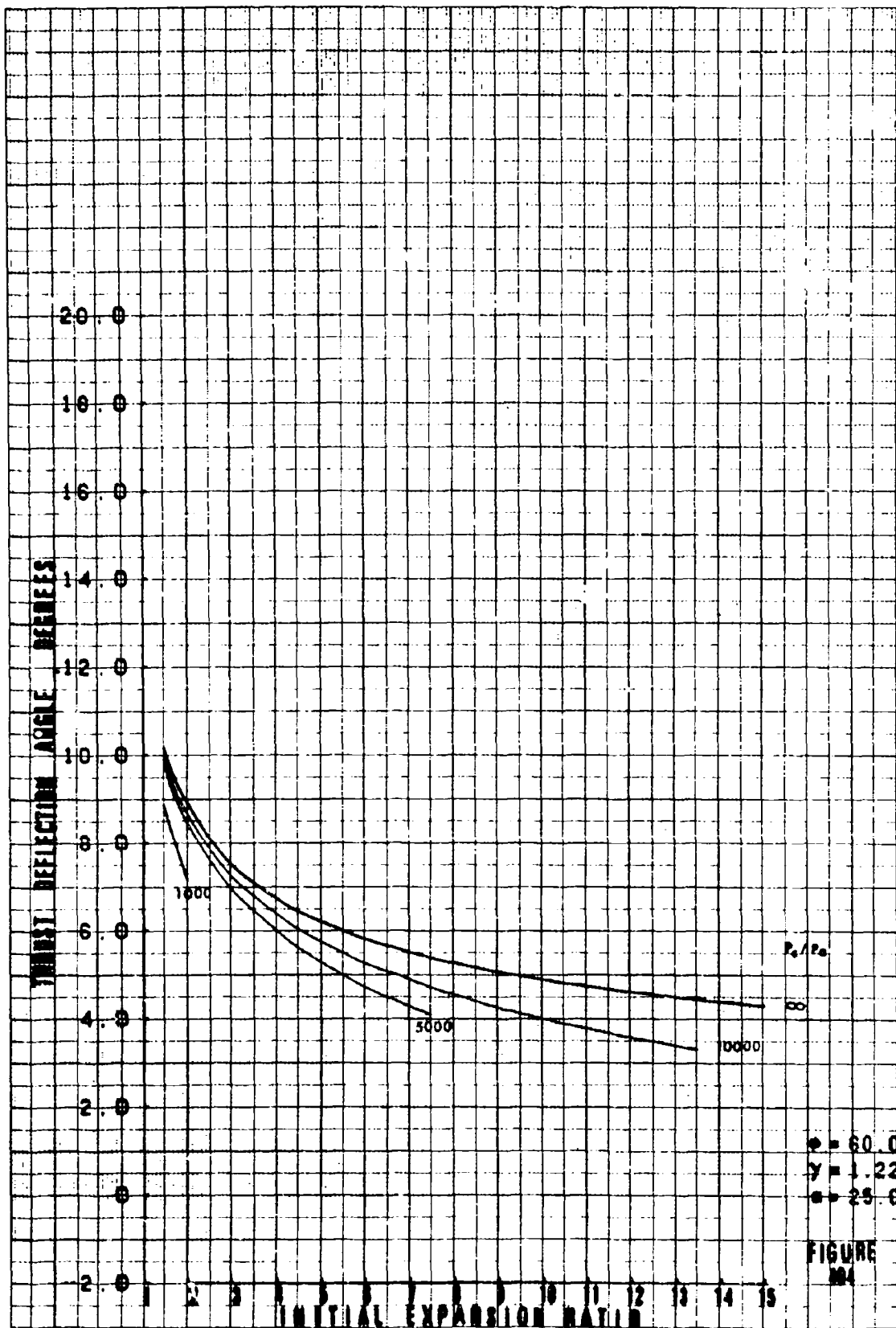


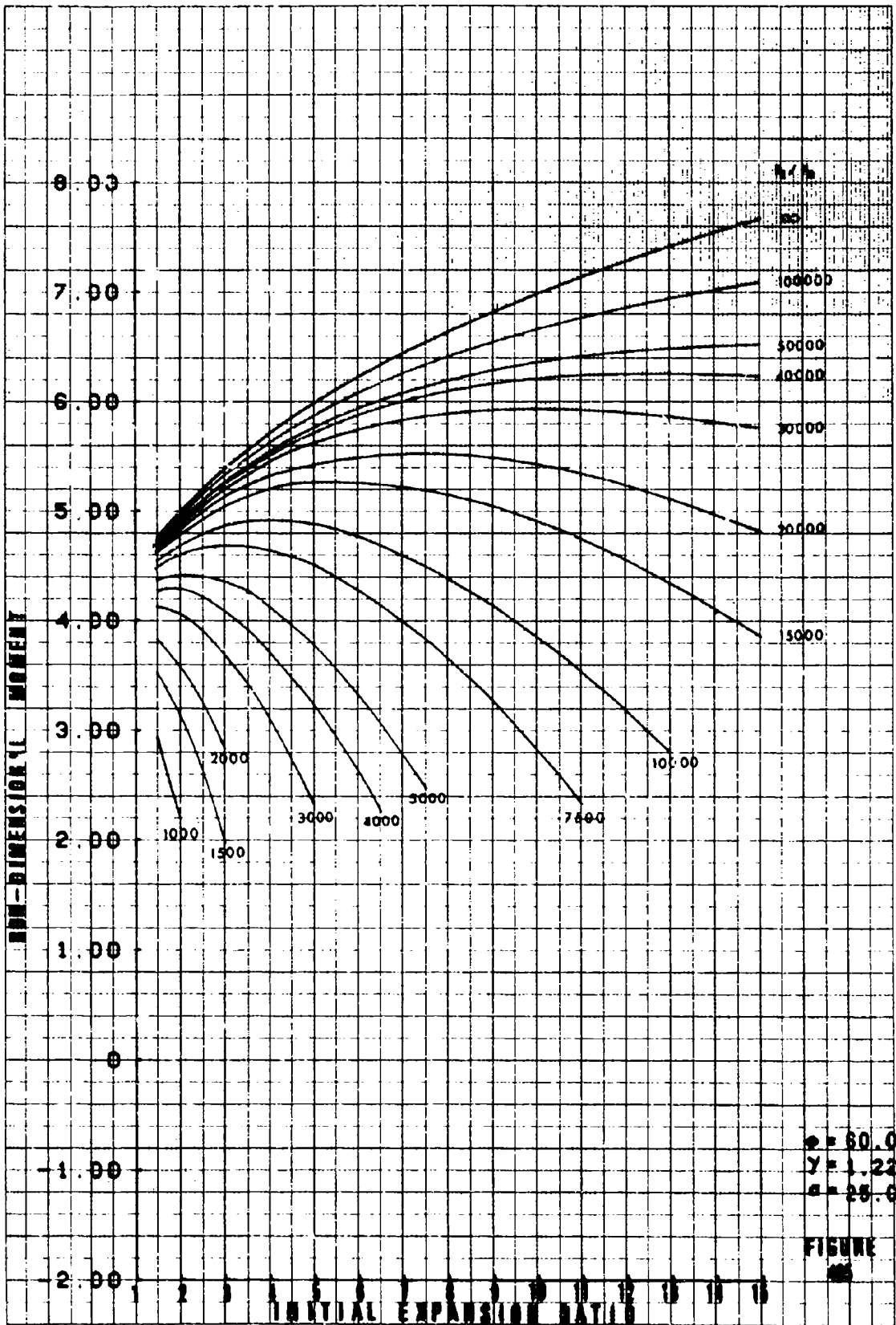


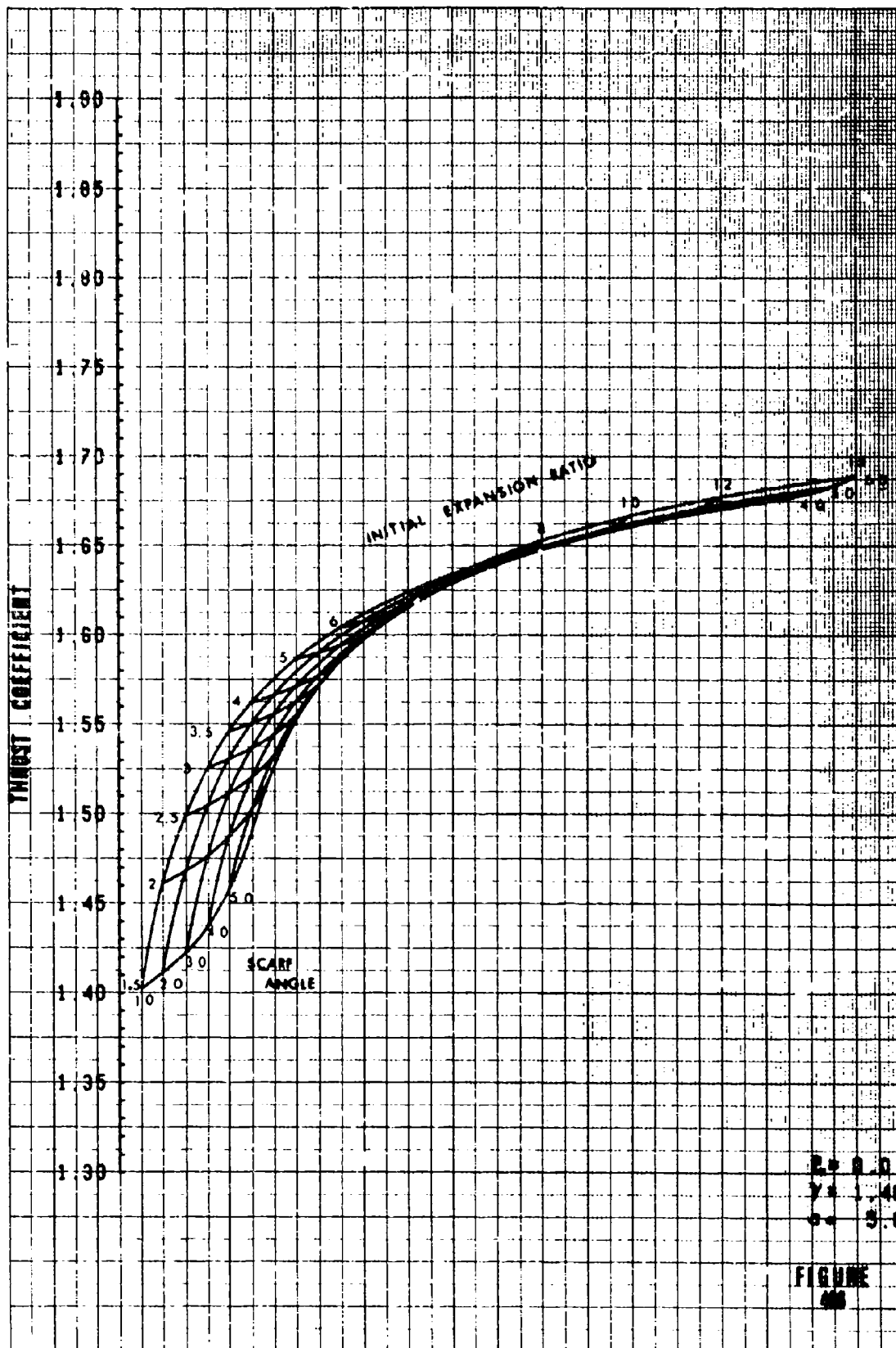


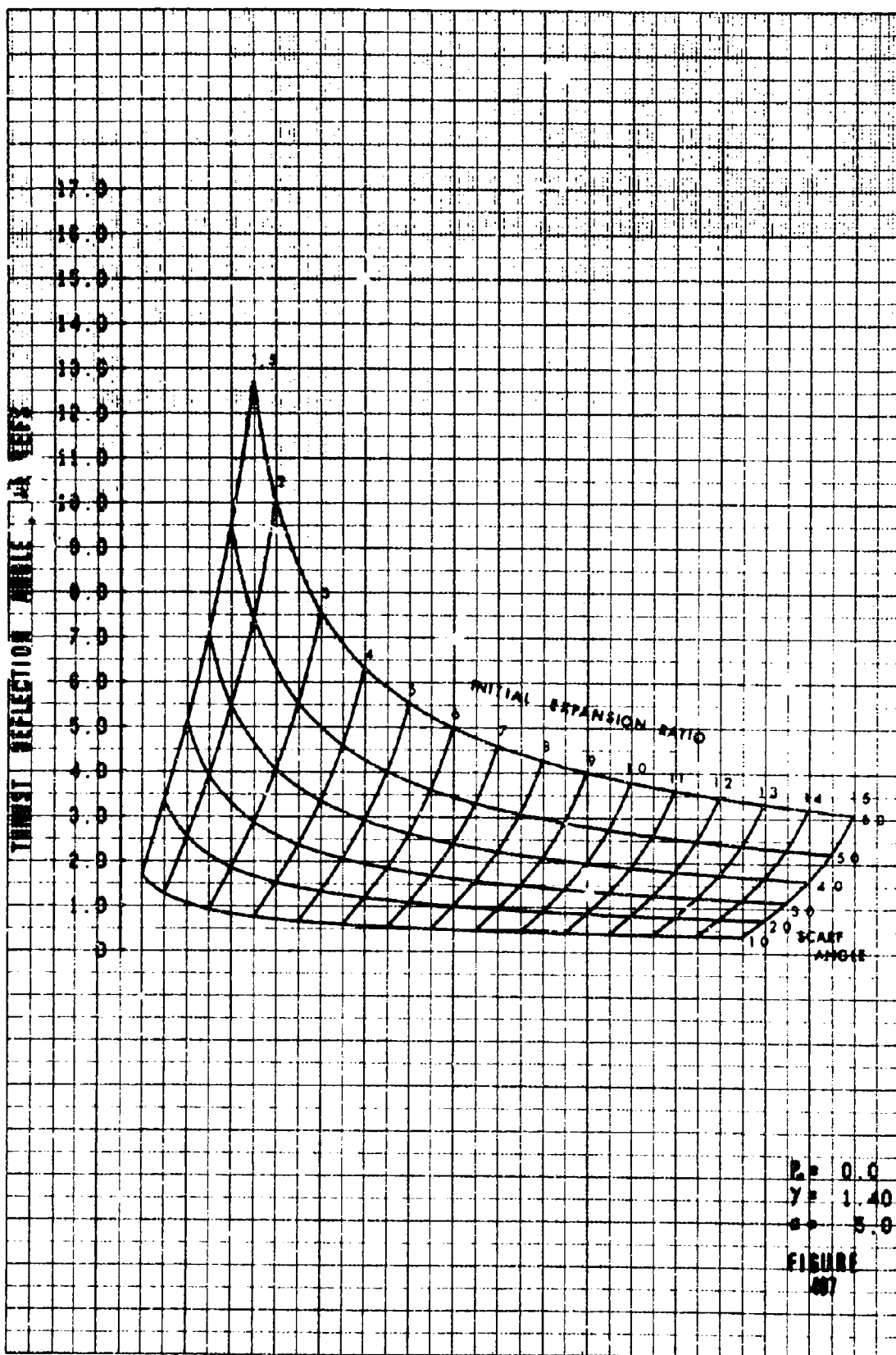


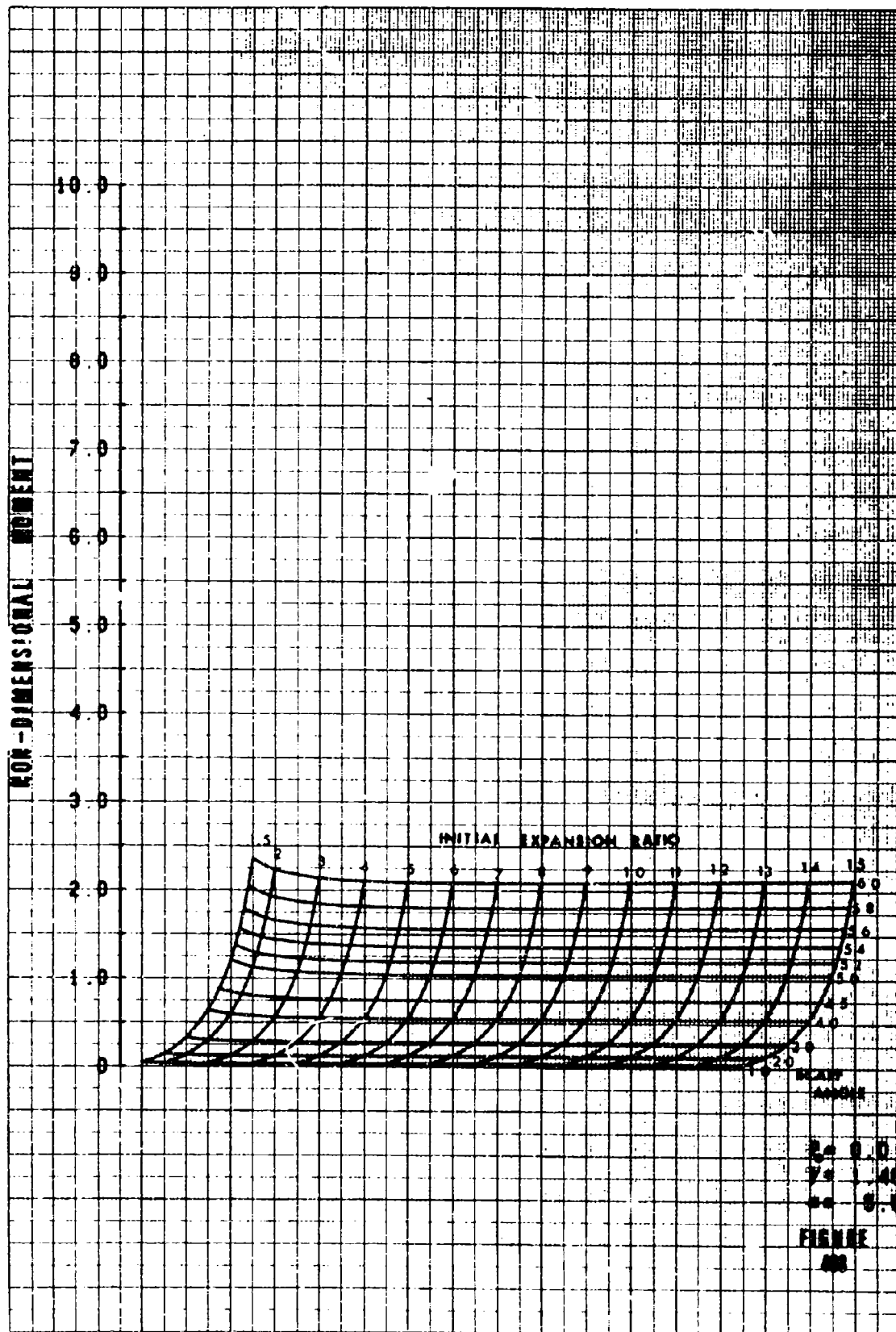


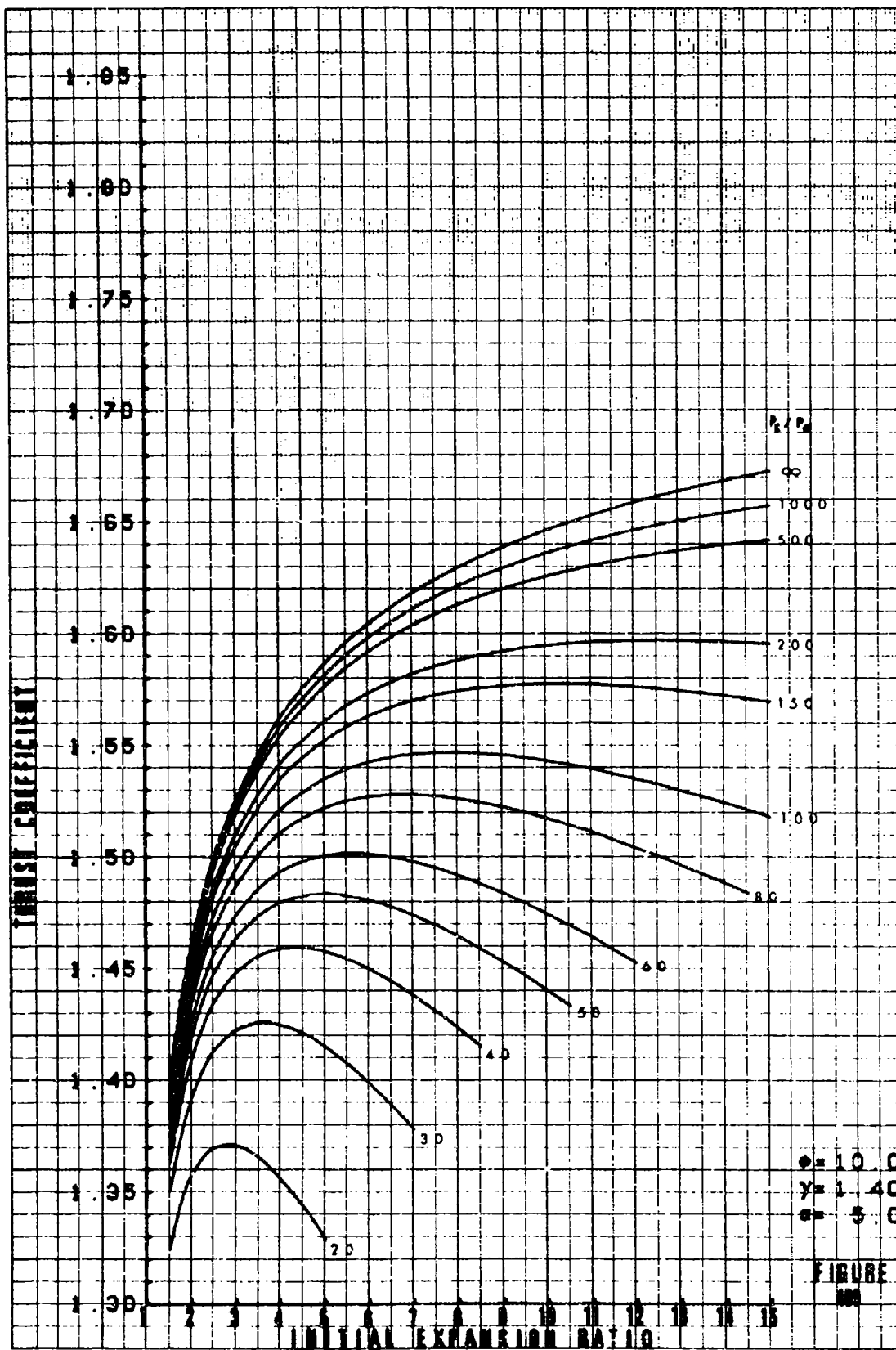


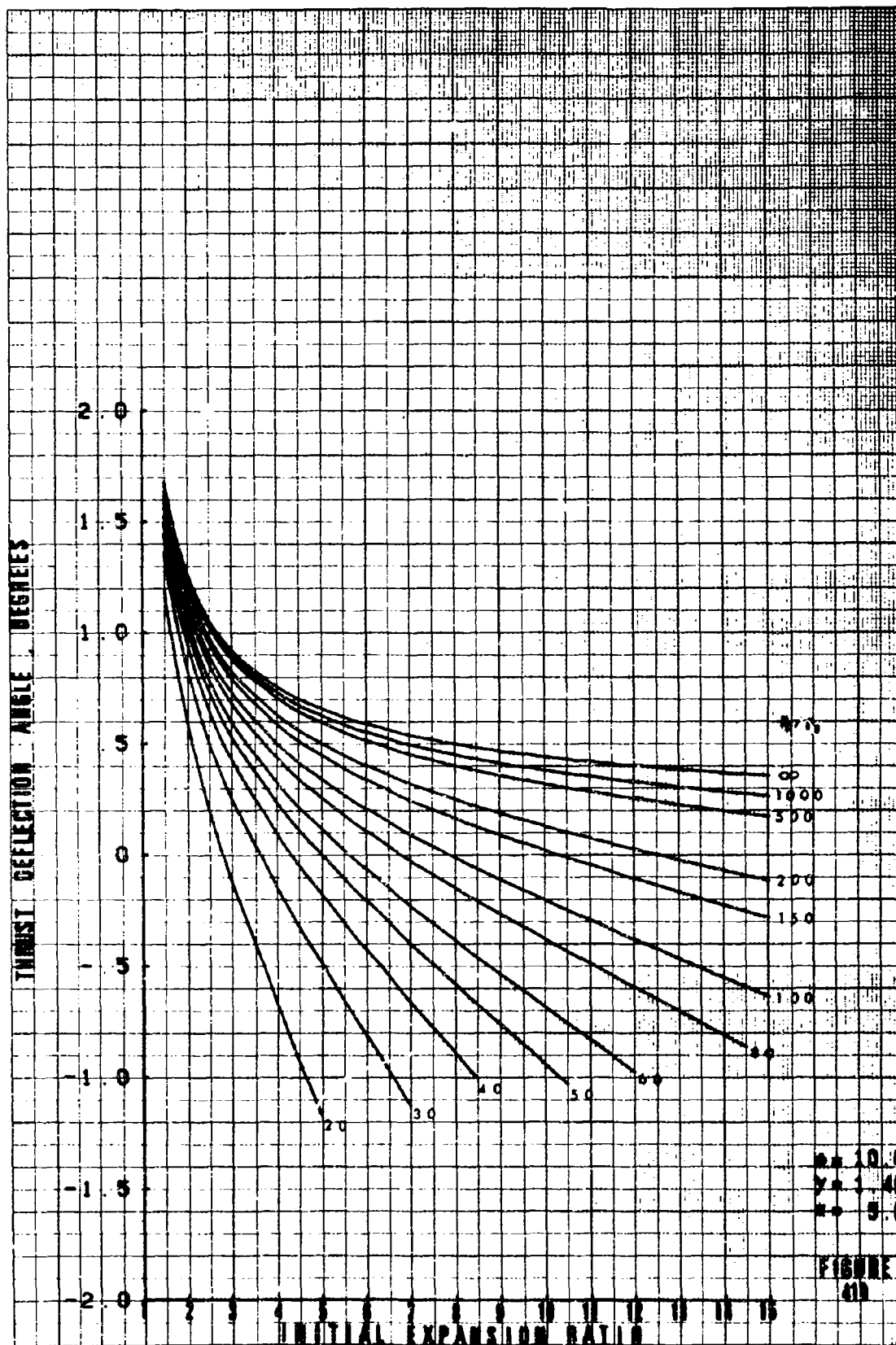






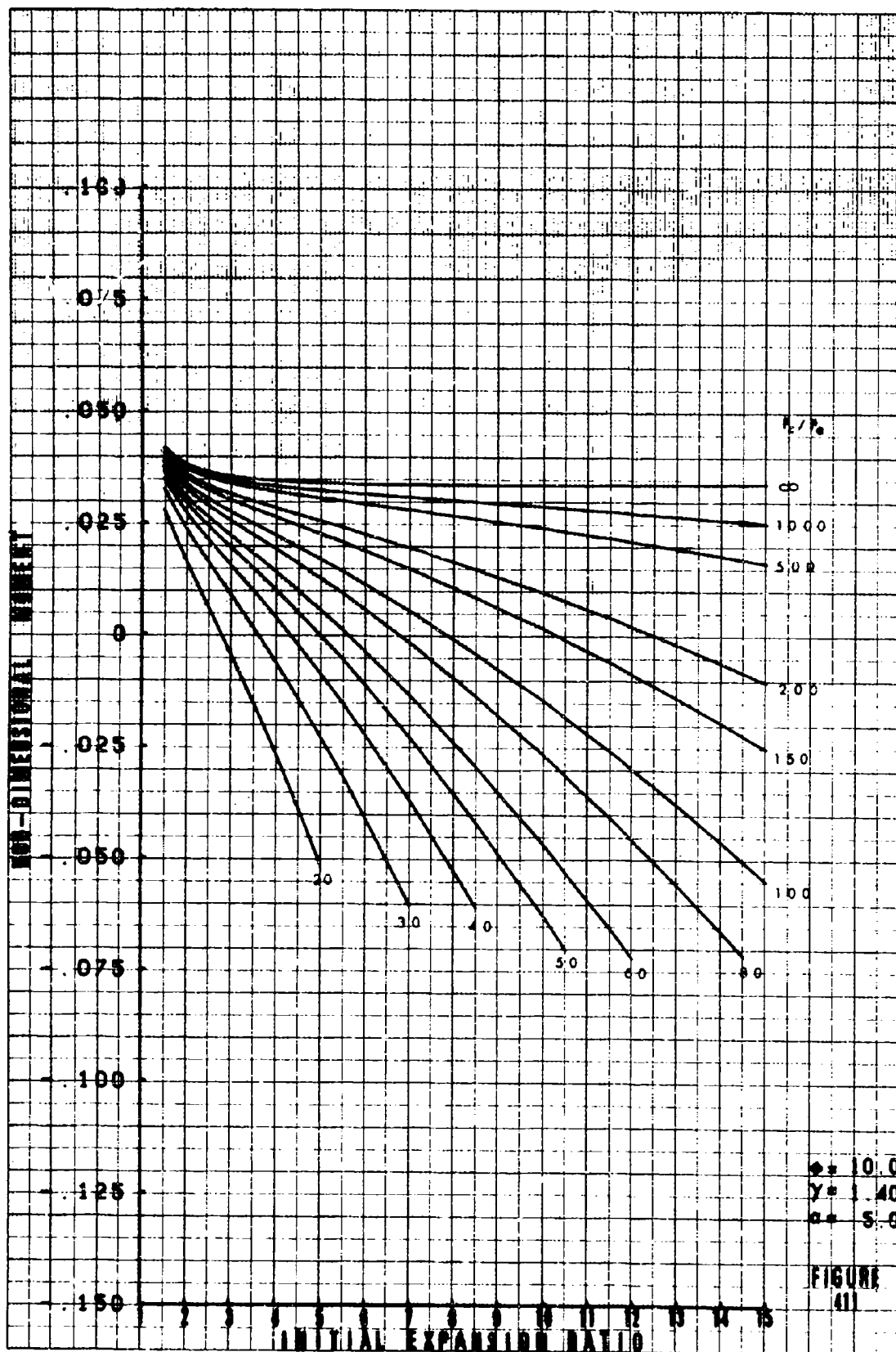


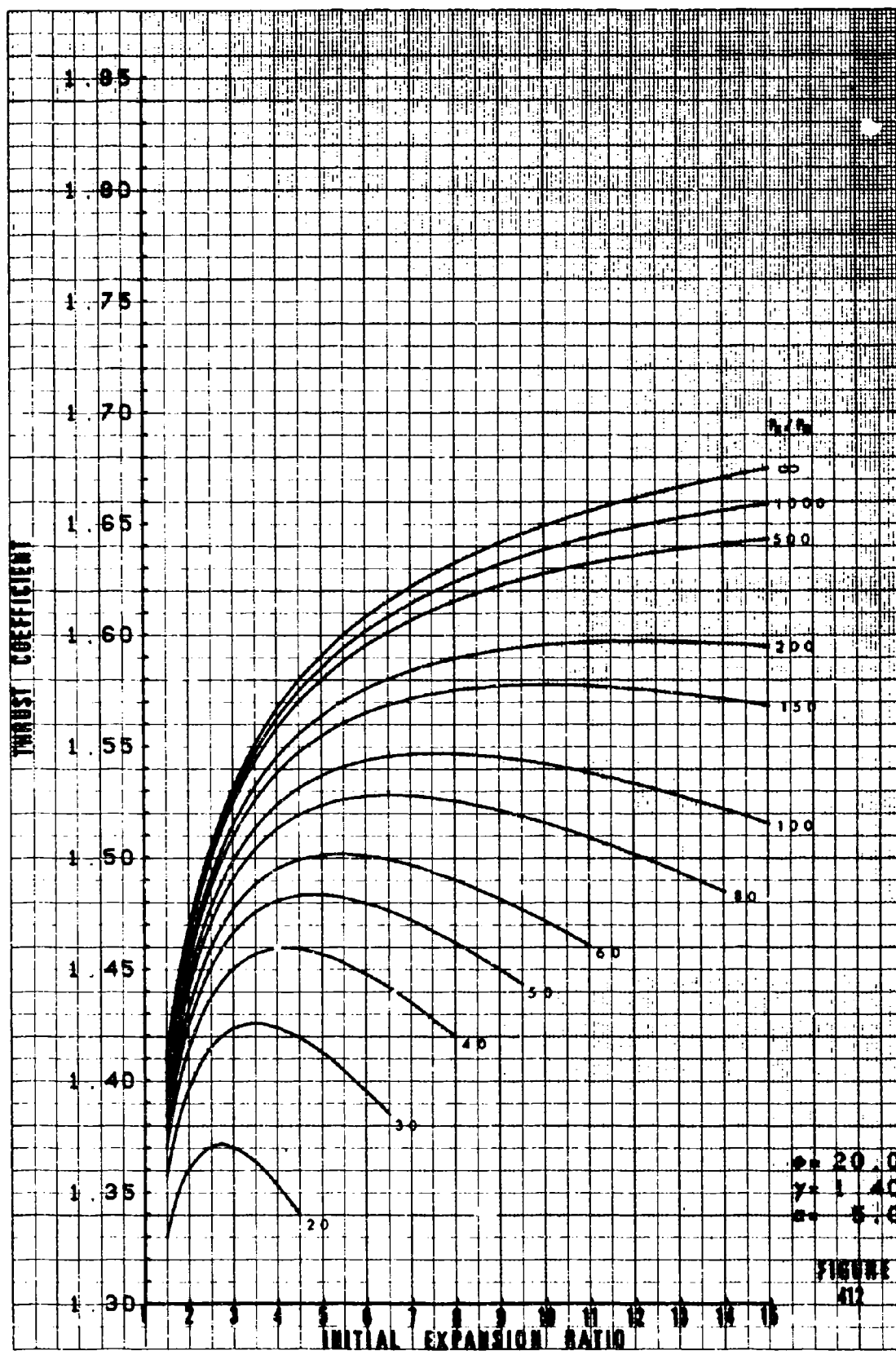




$M_{\infty} = 10.0$
 $M_{\infty} = 5.0$
 $M_{\infty} = 2.0$

FIGURE 413





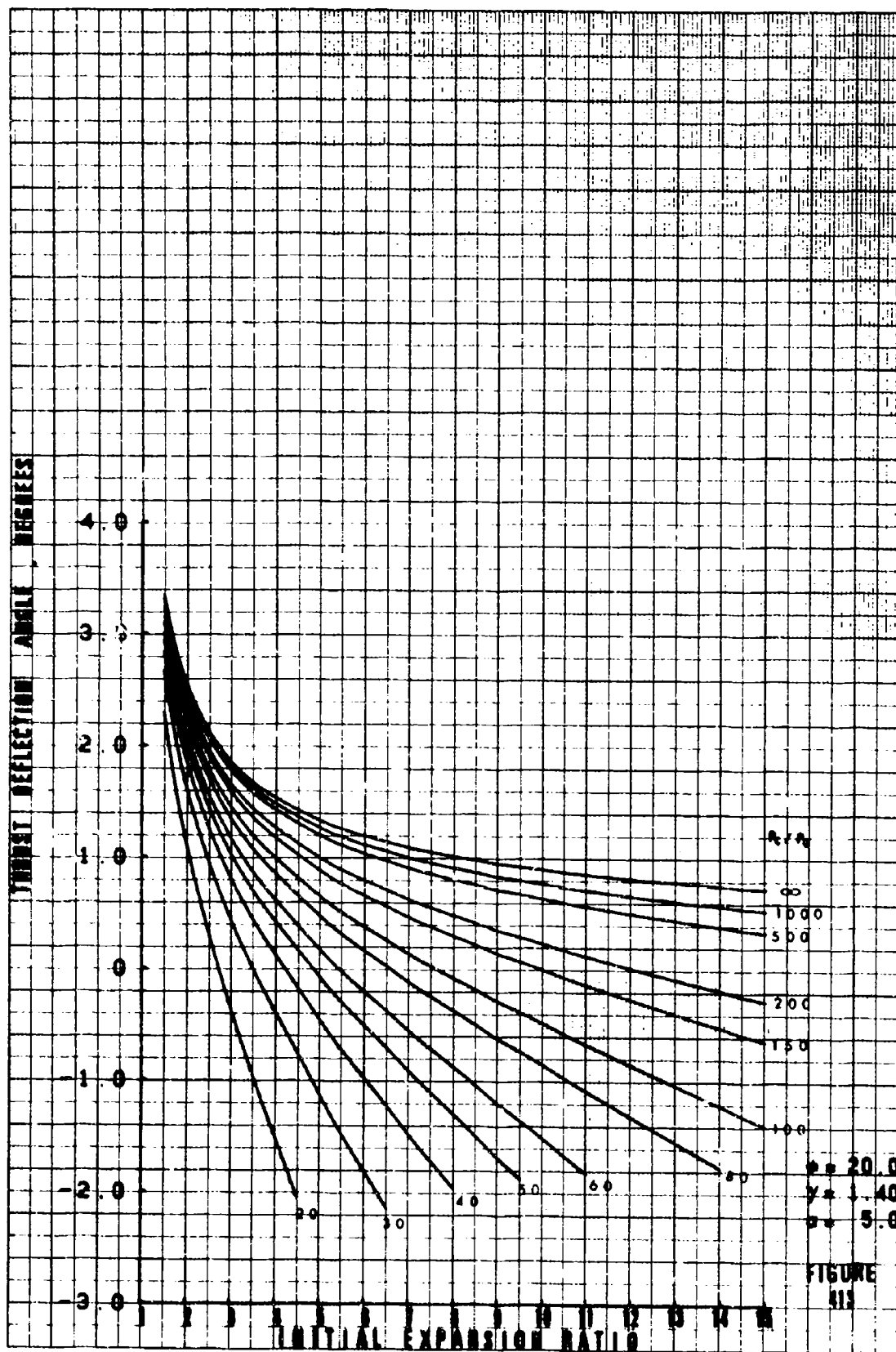
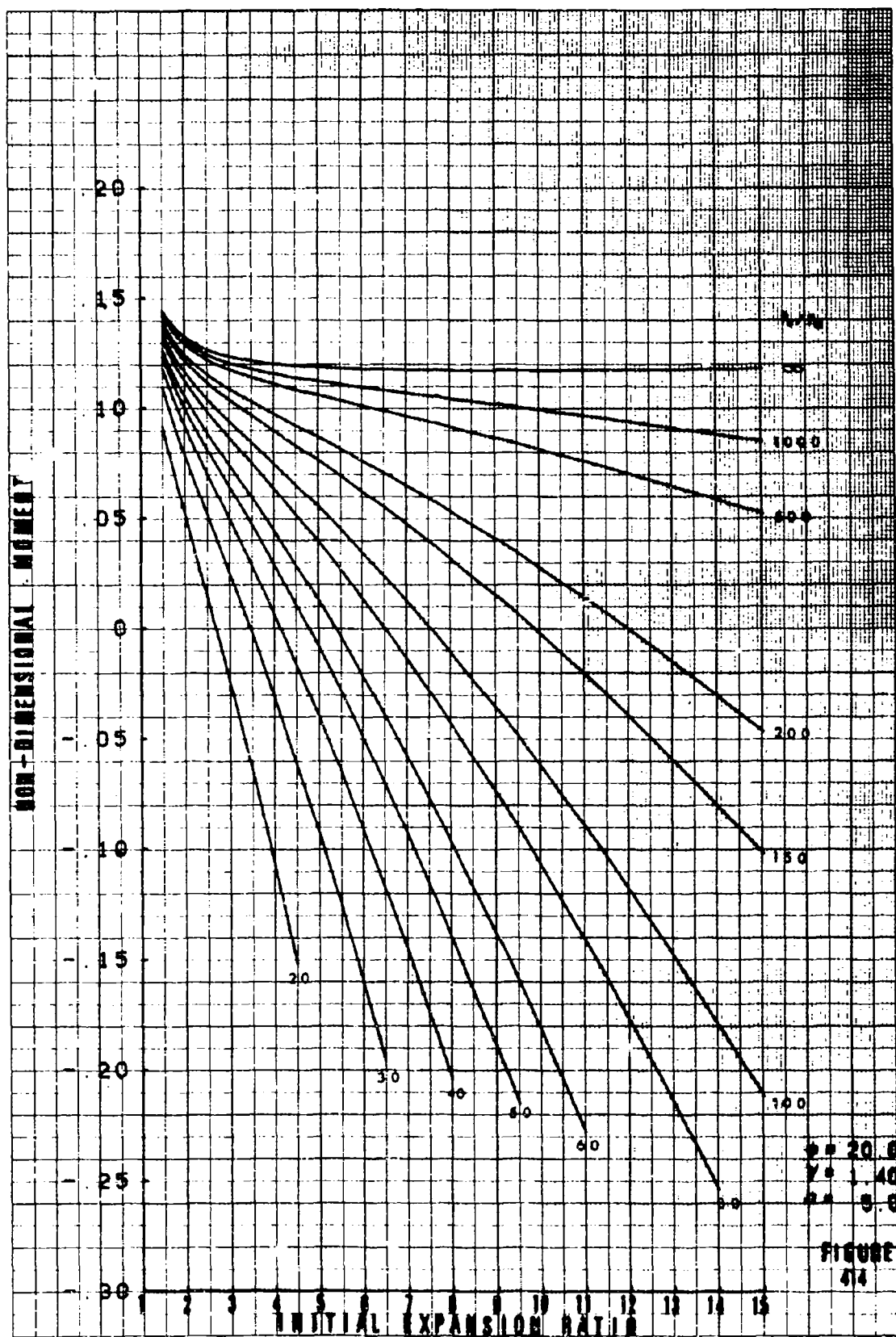
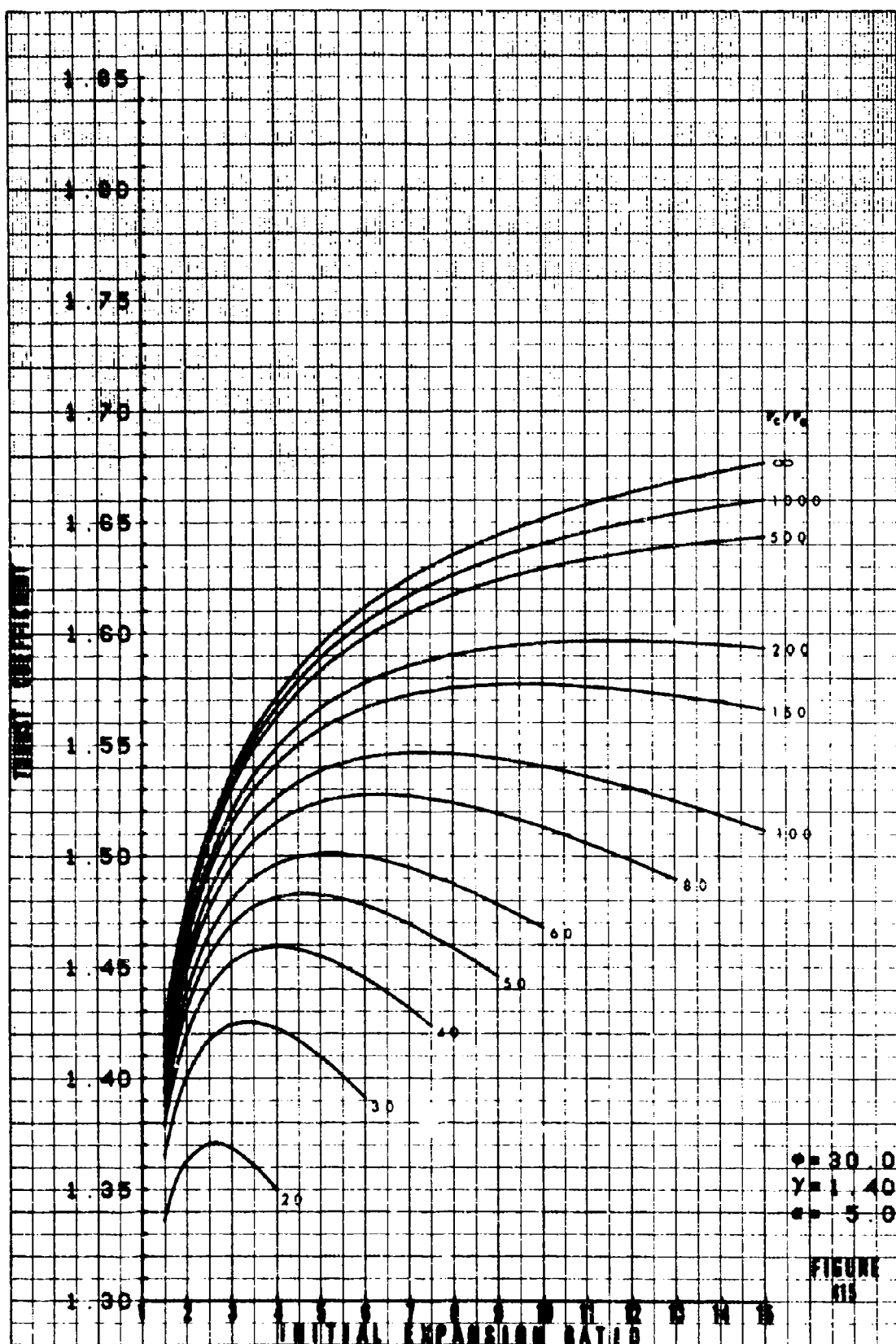
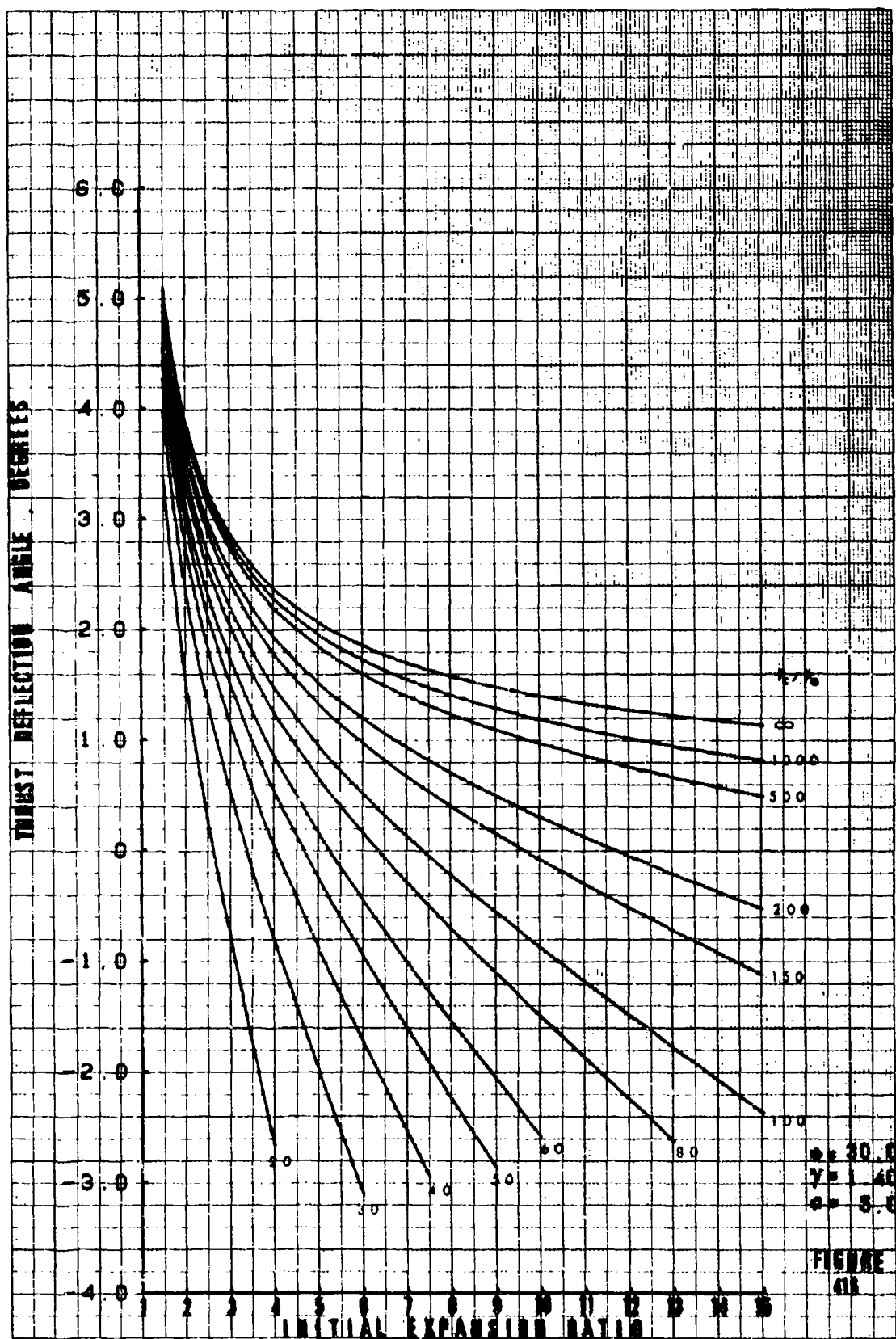
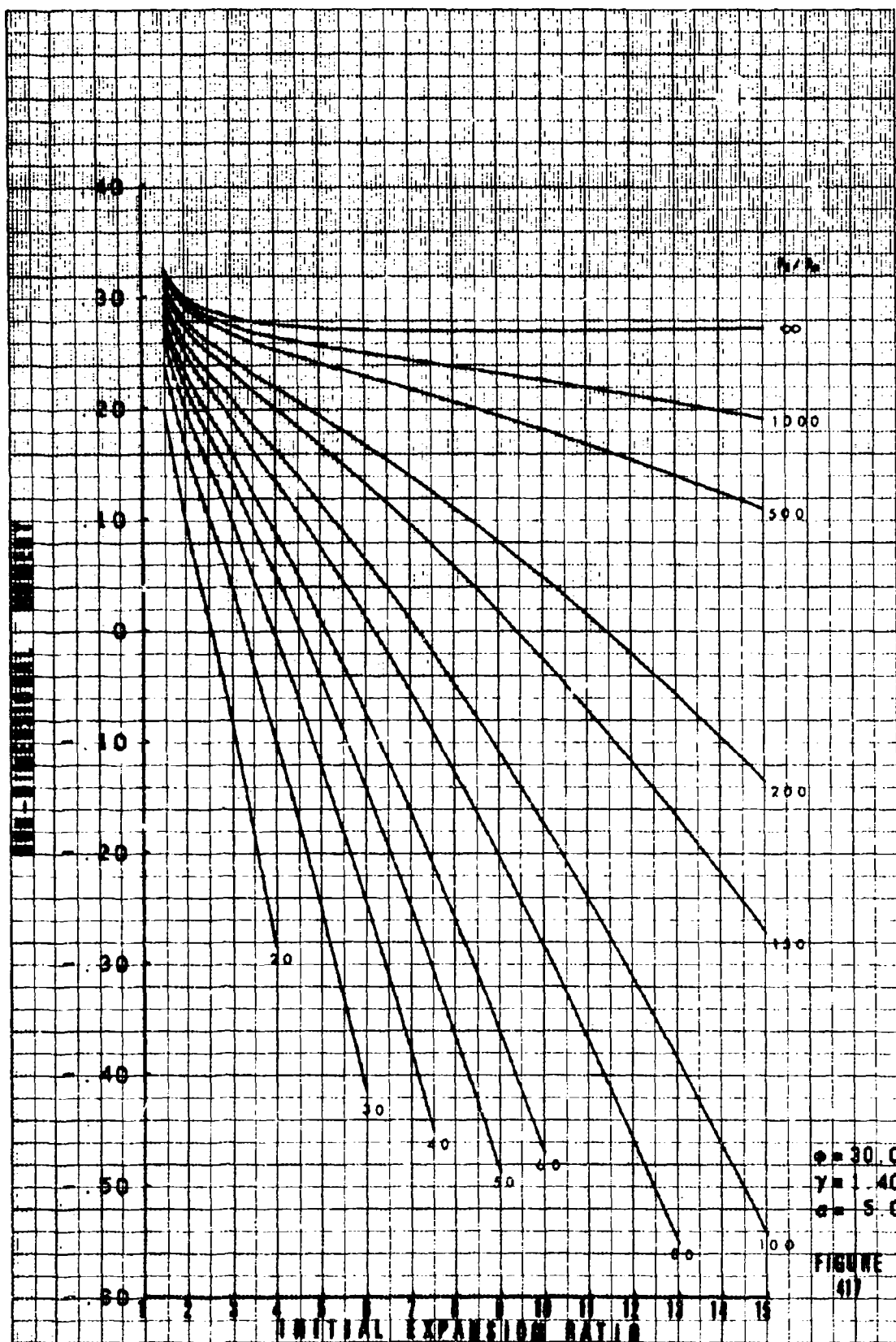


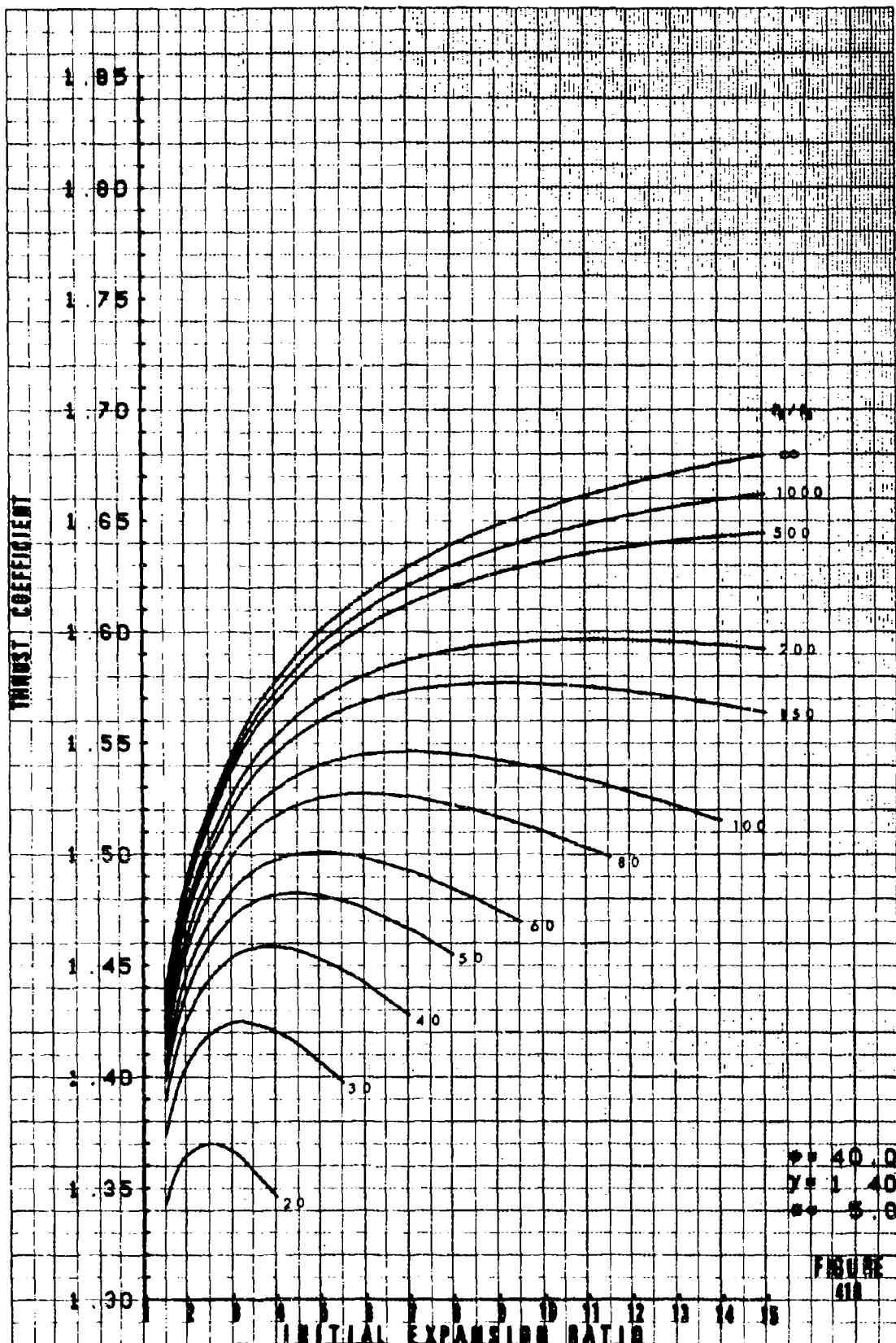
FIGURE
415

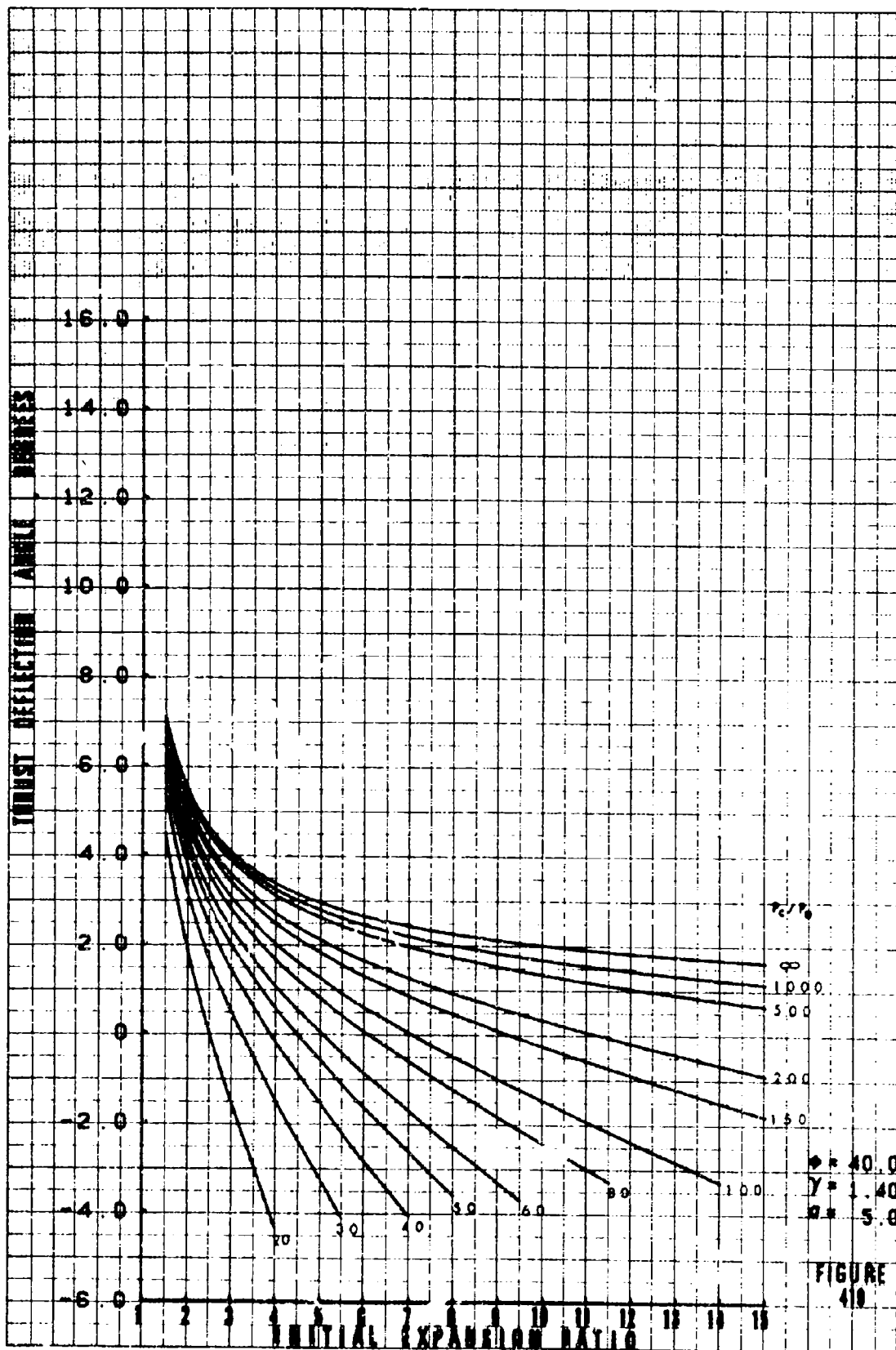


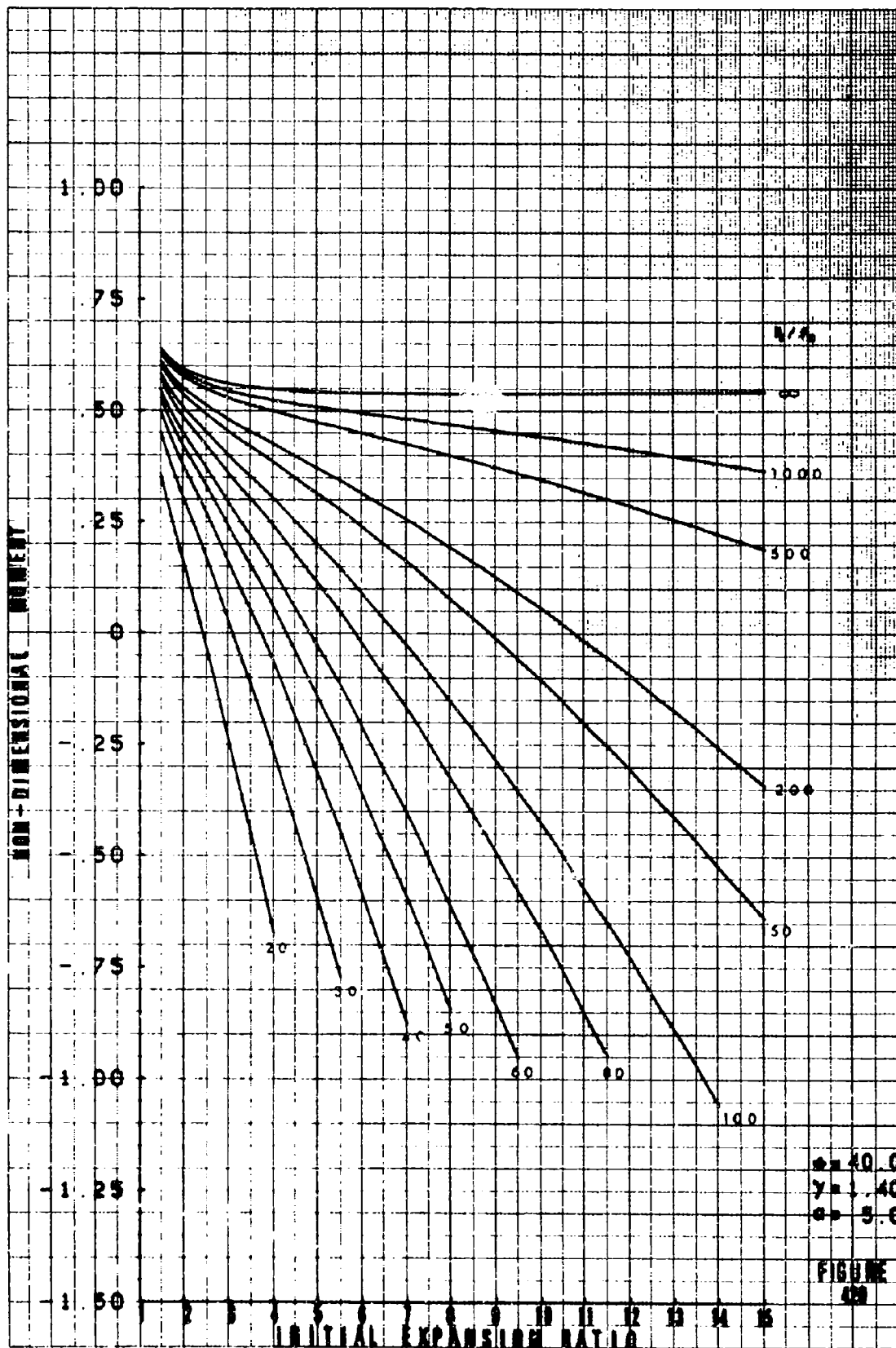


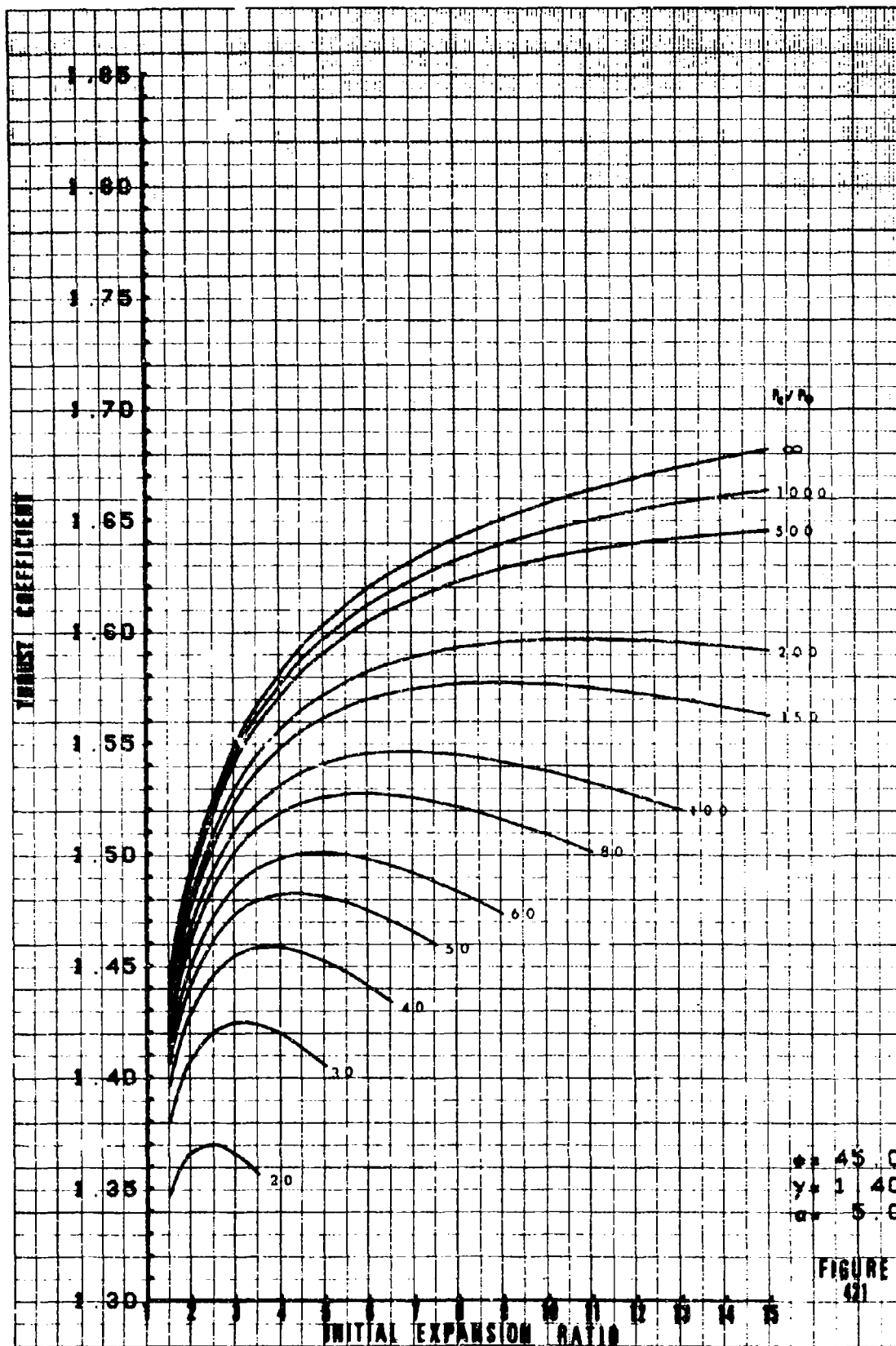


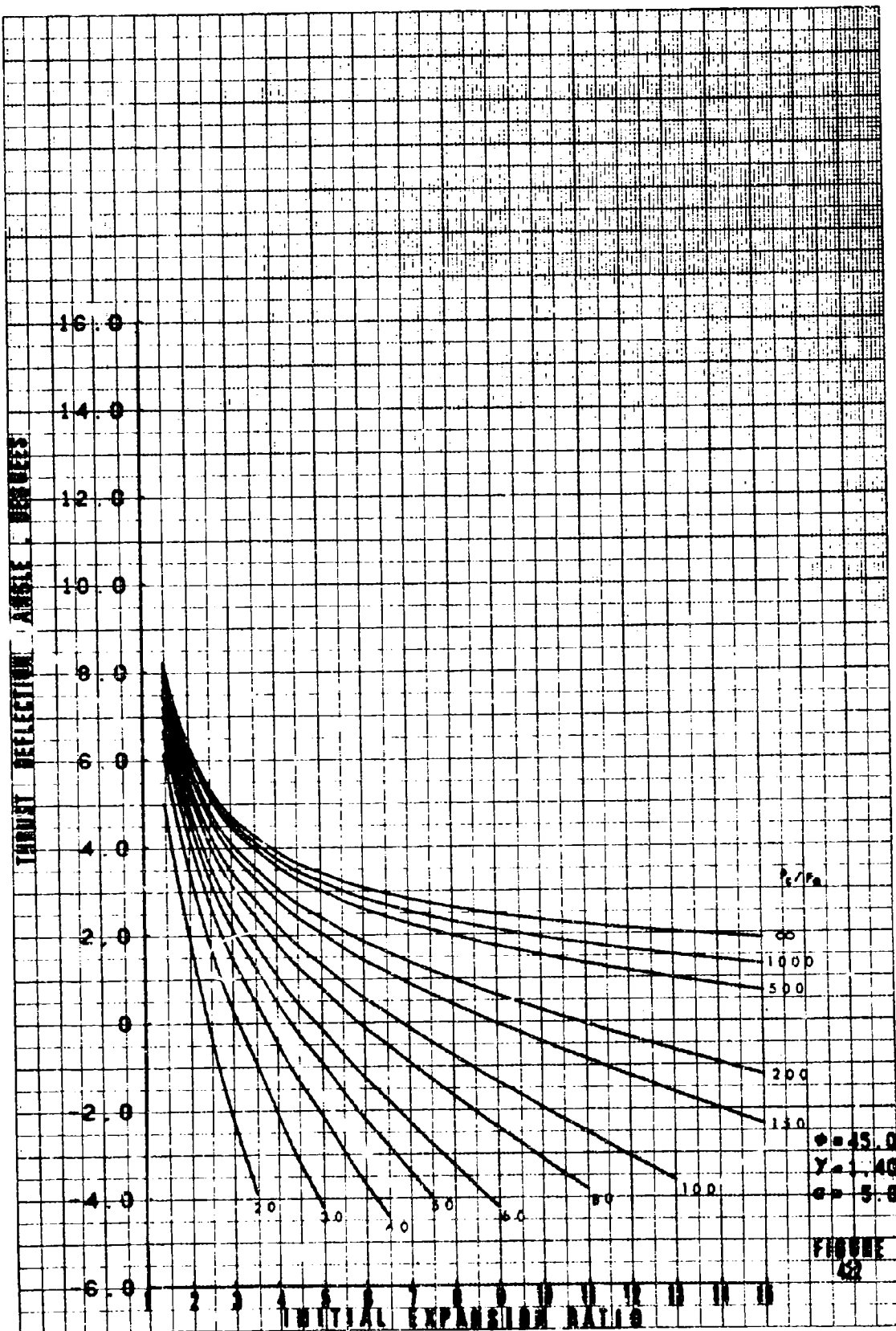












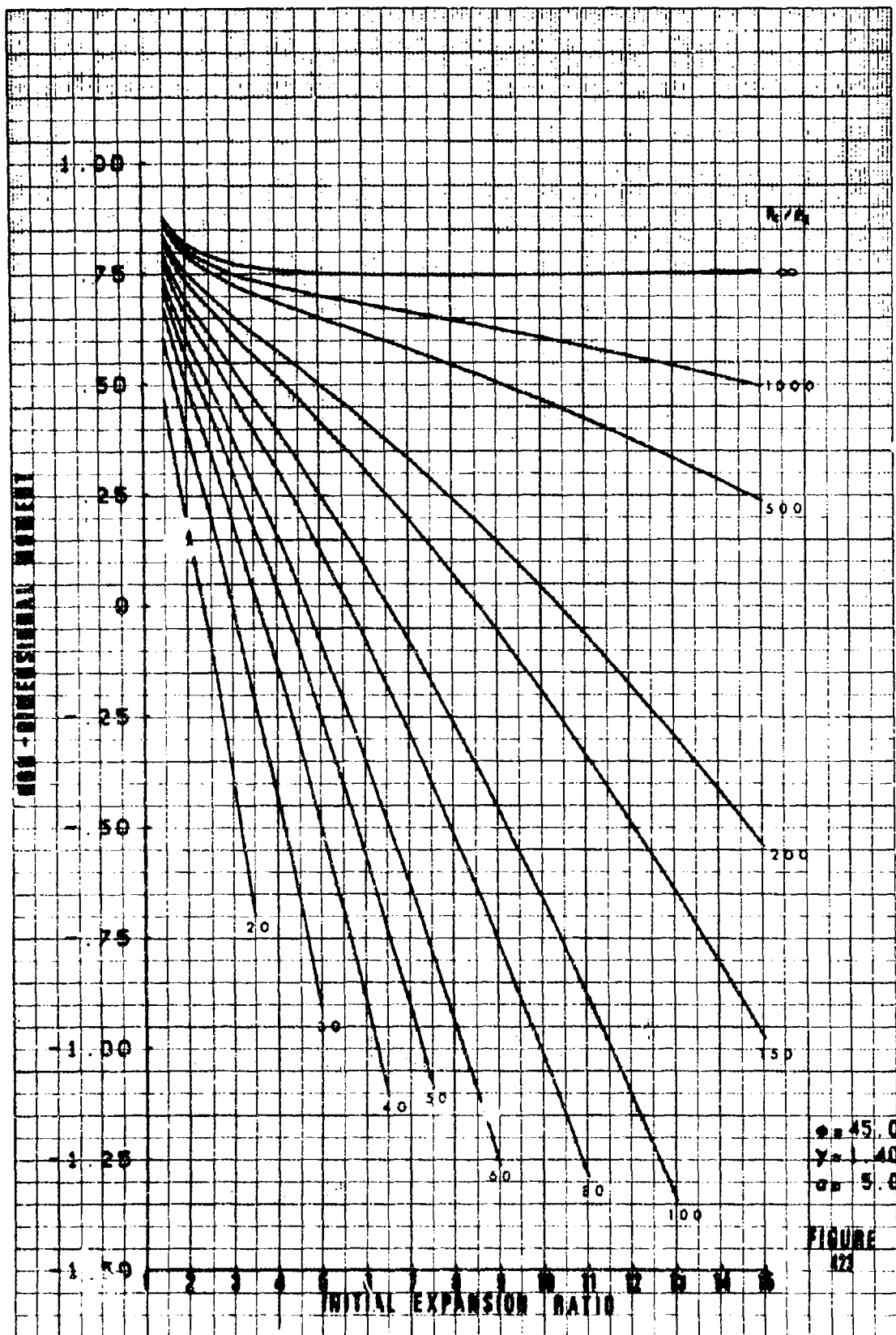


FIGURE 421

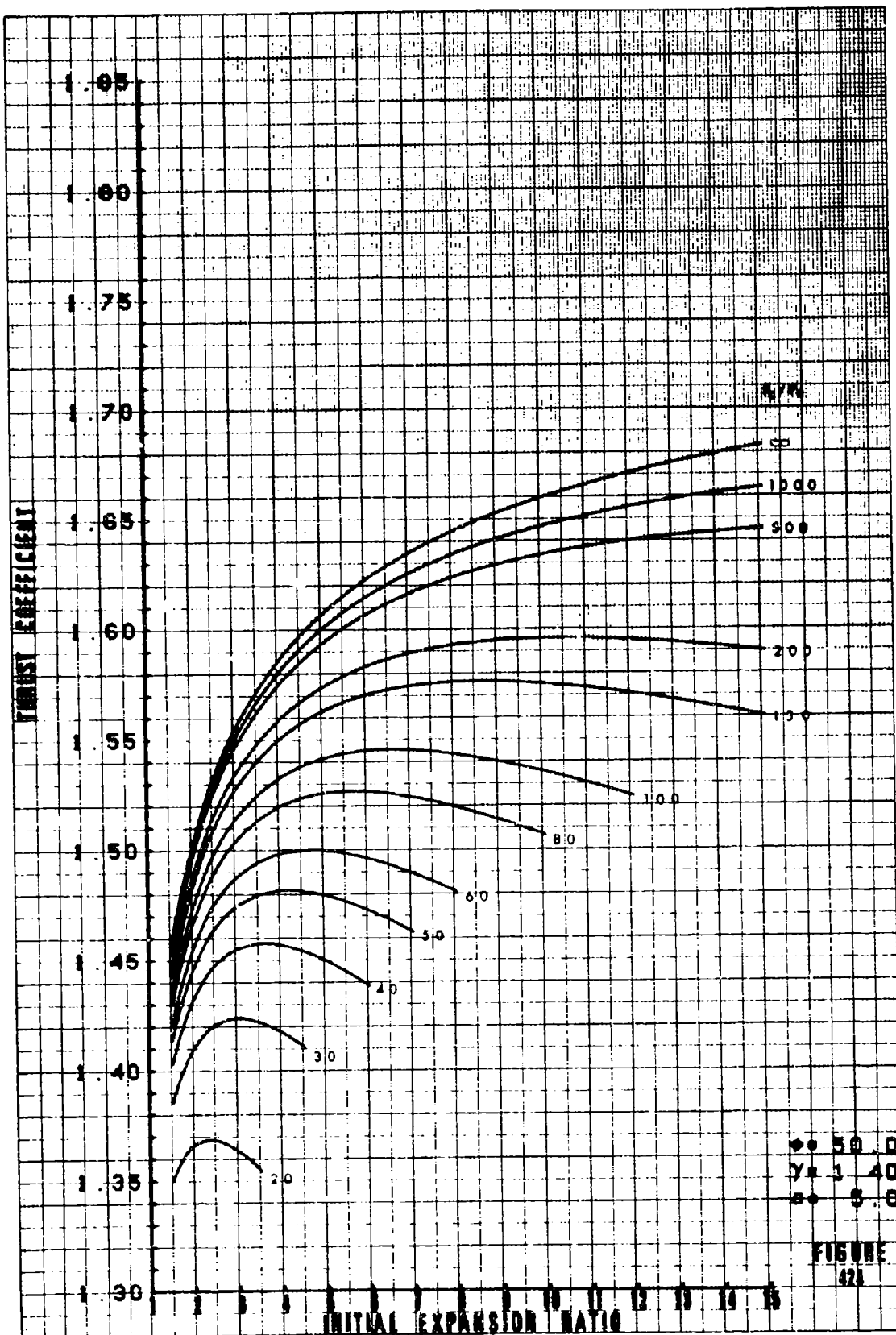
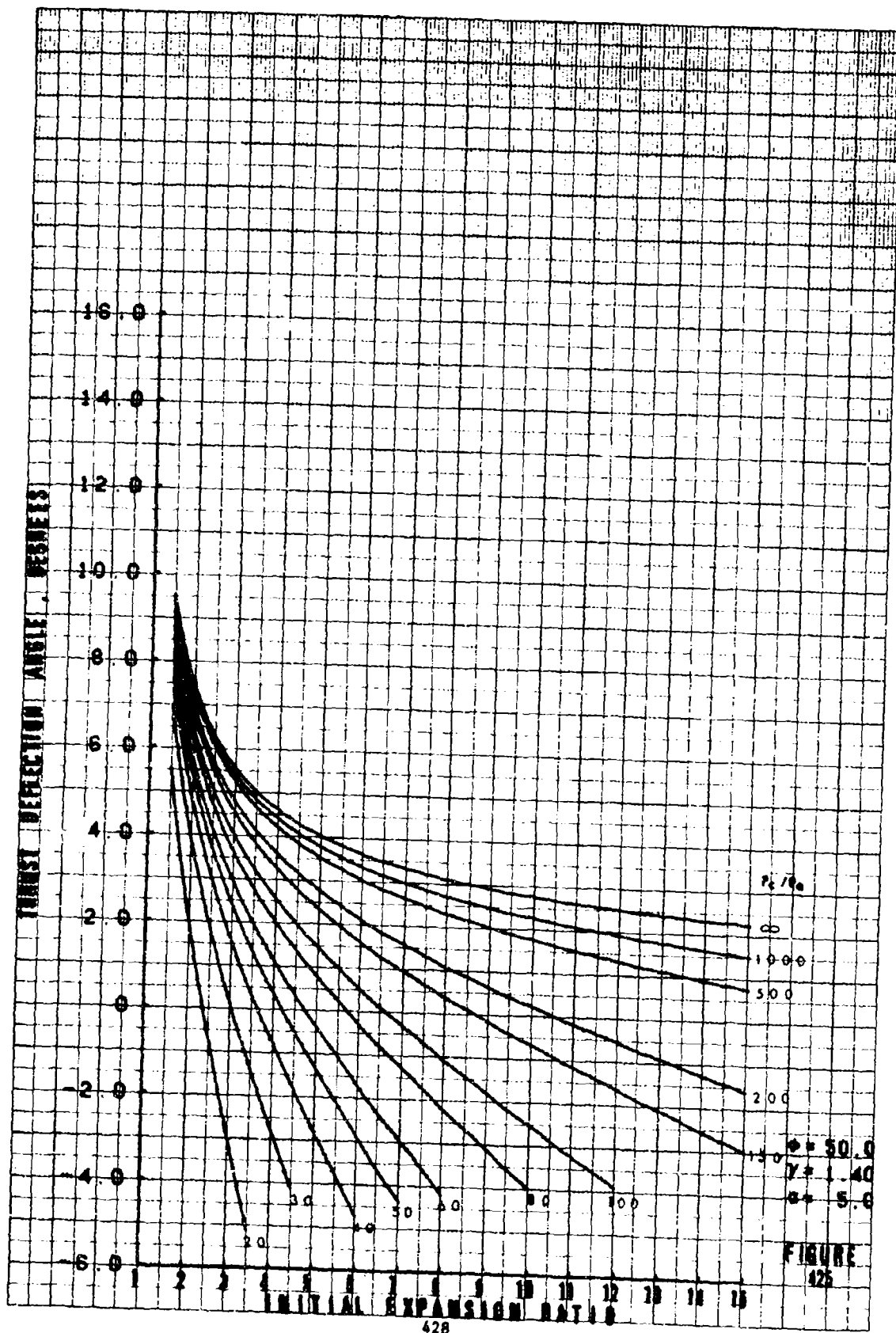
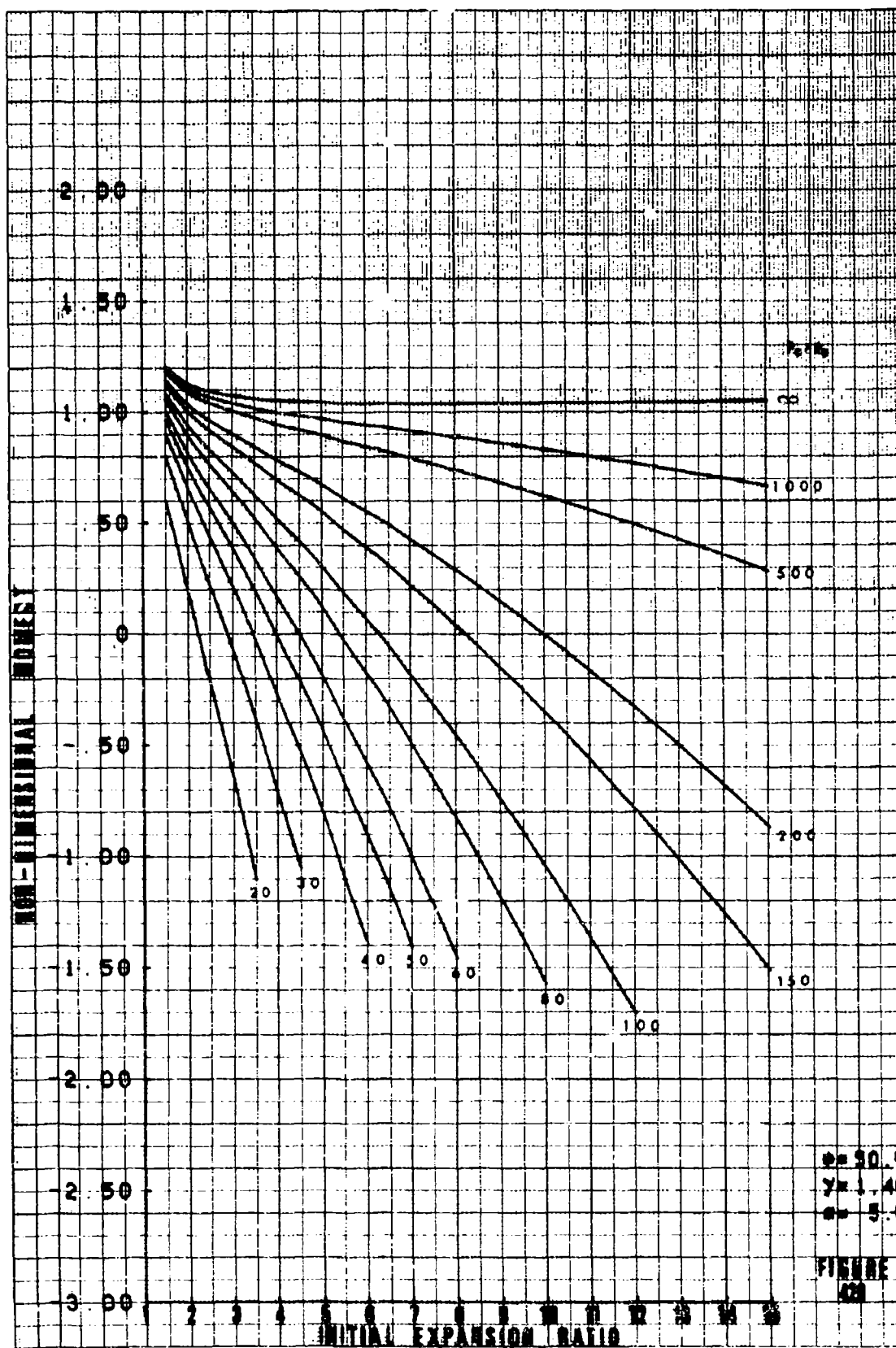
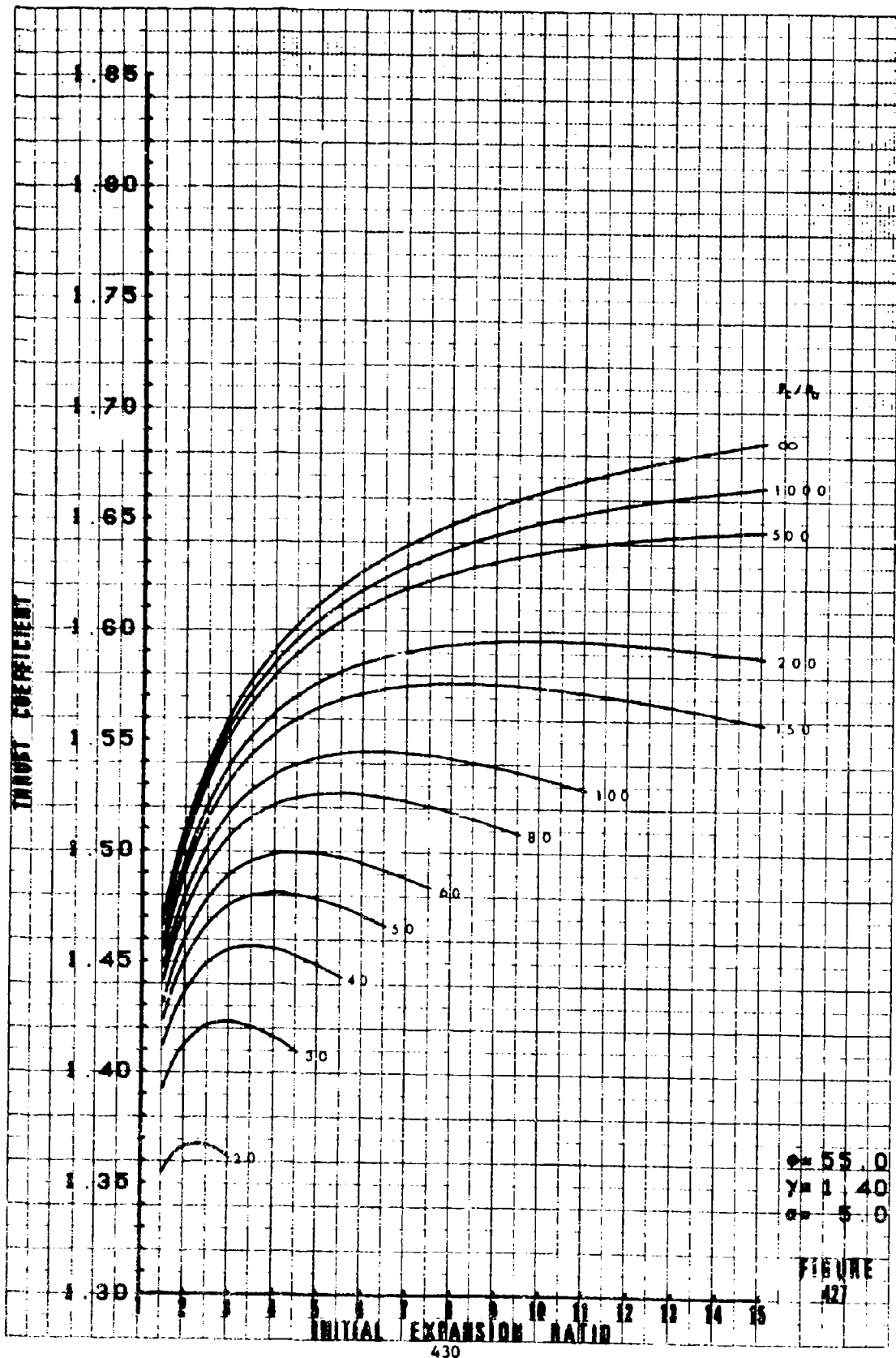
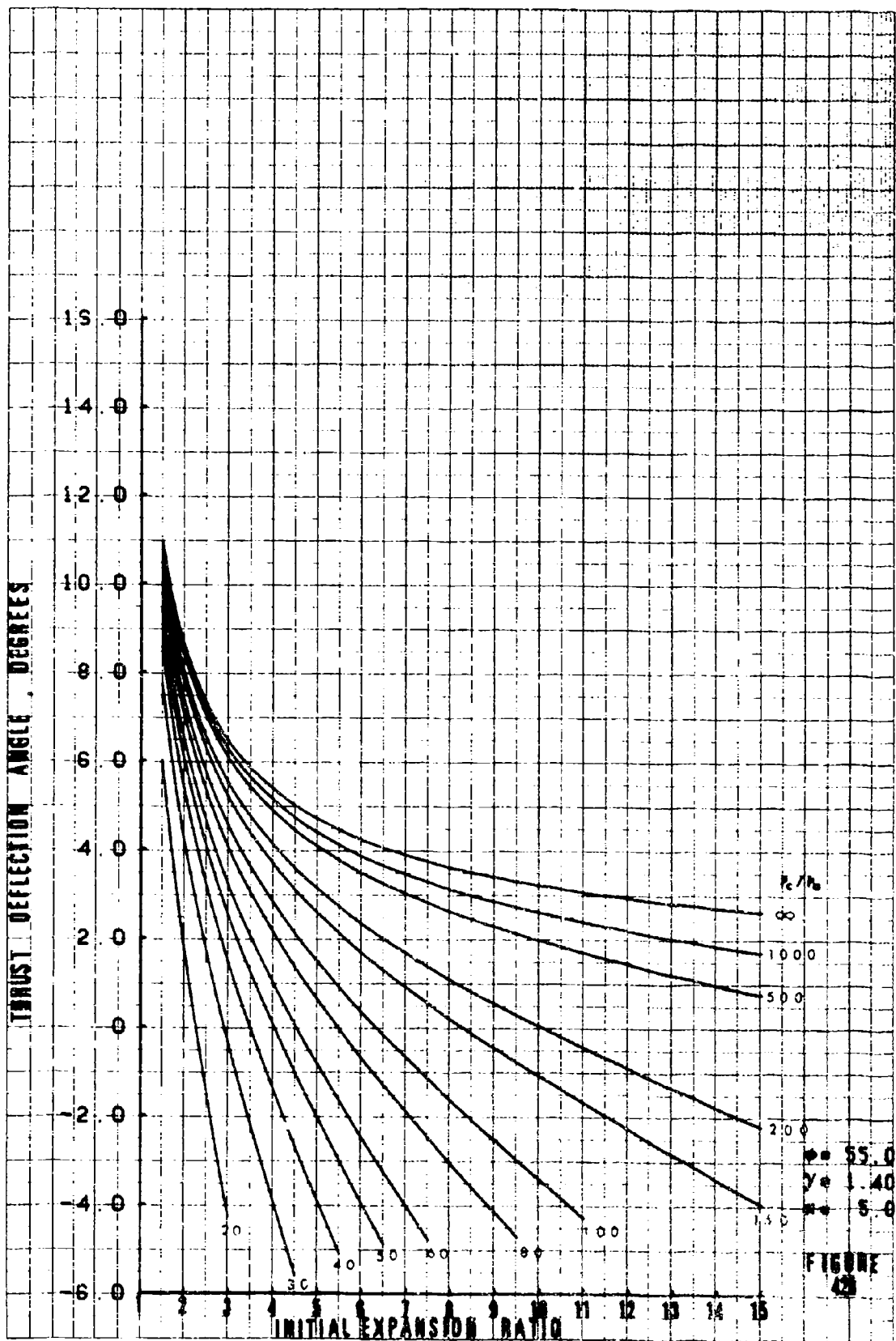


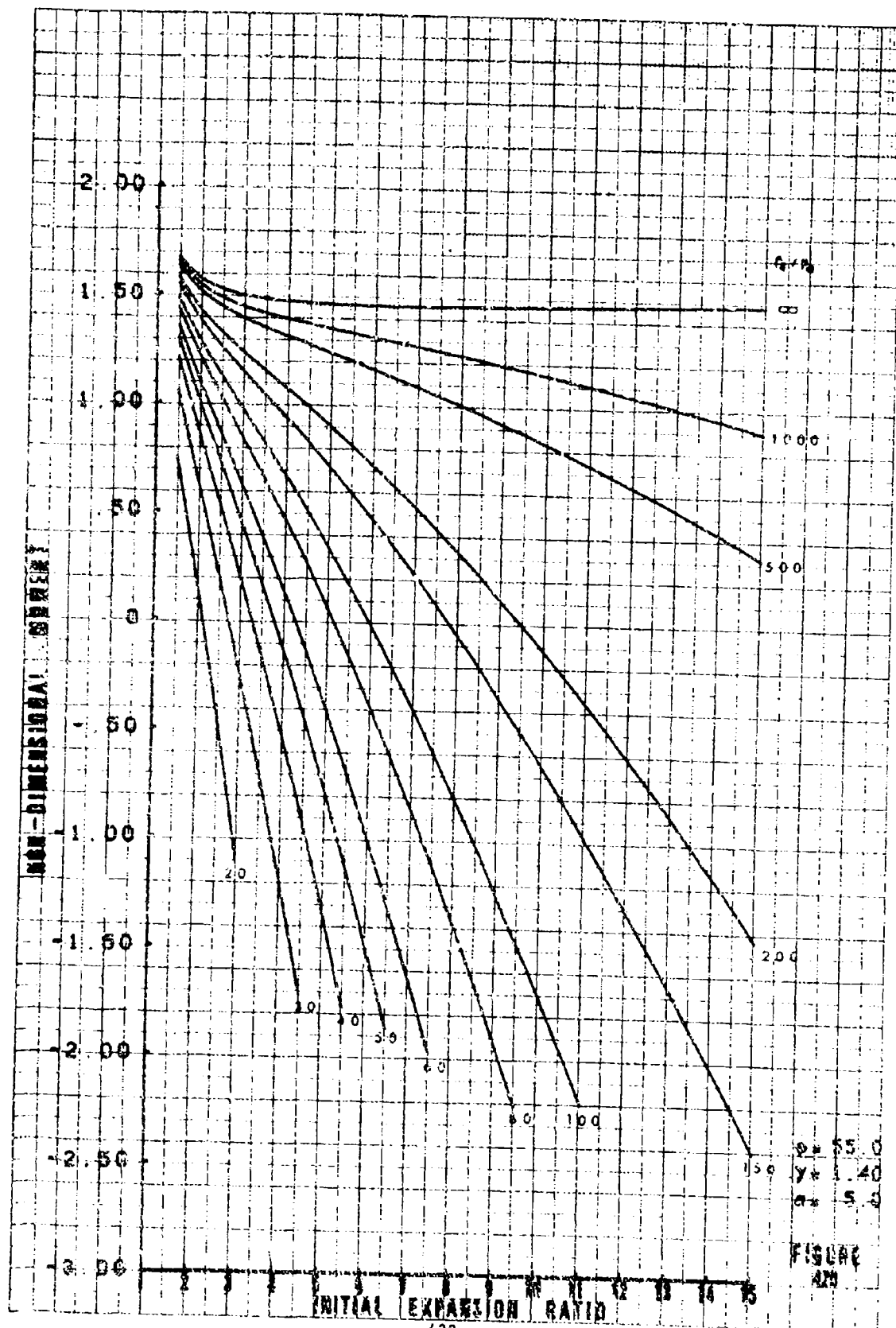
FIGURE 423

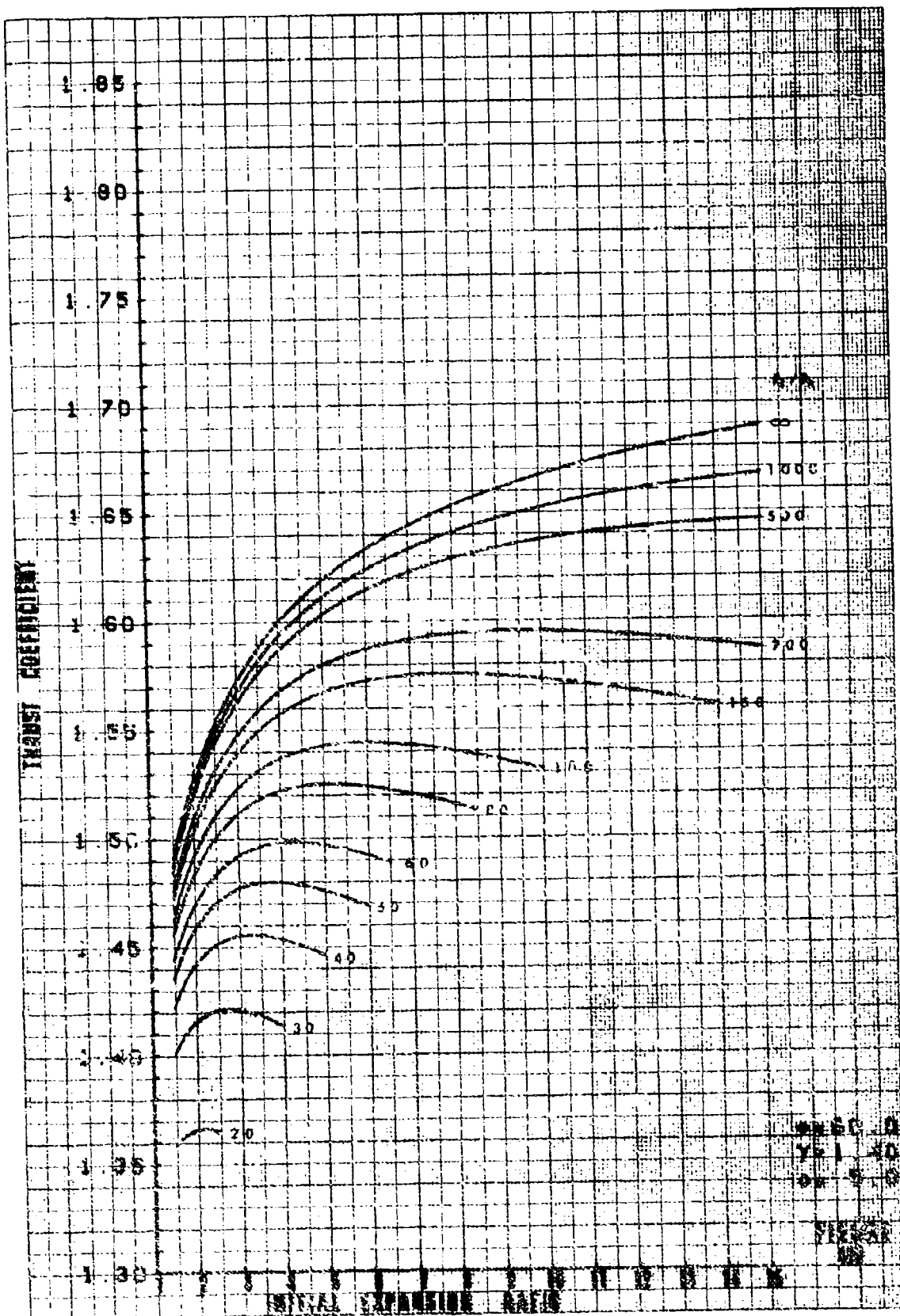












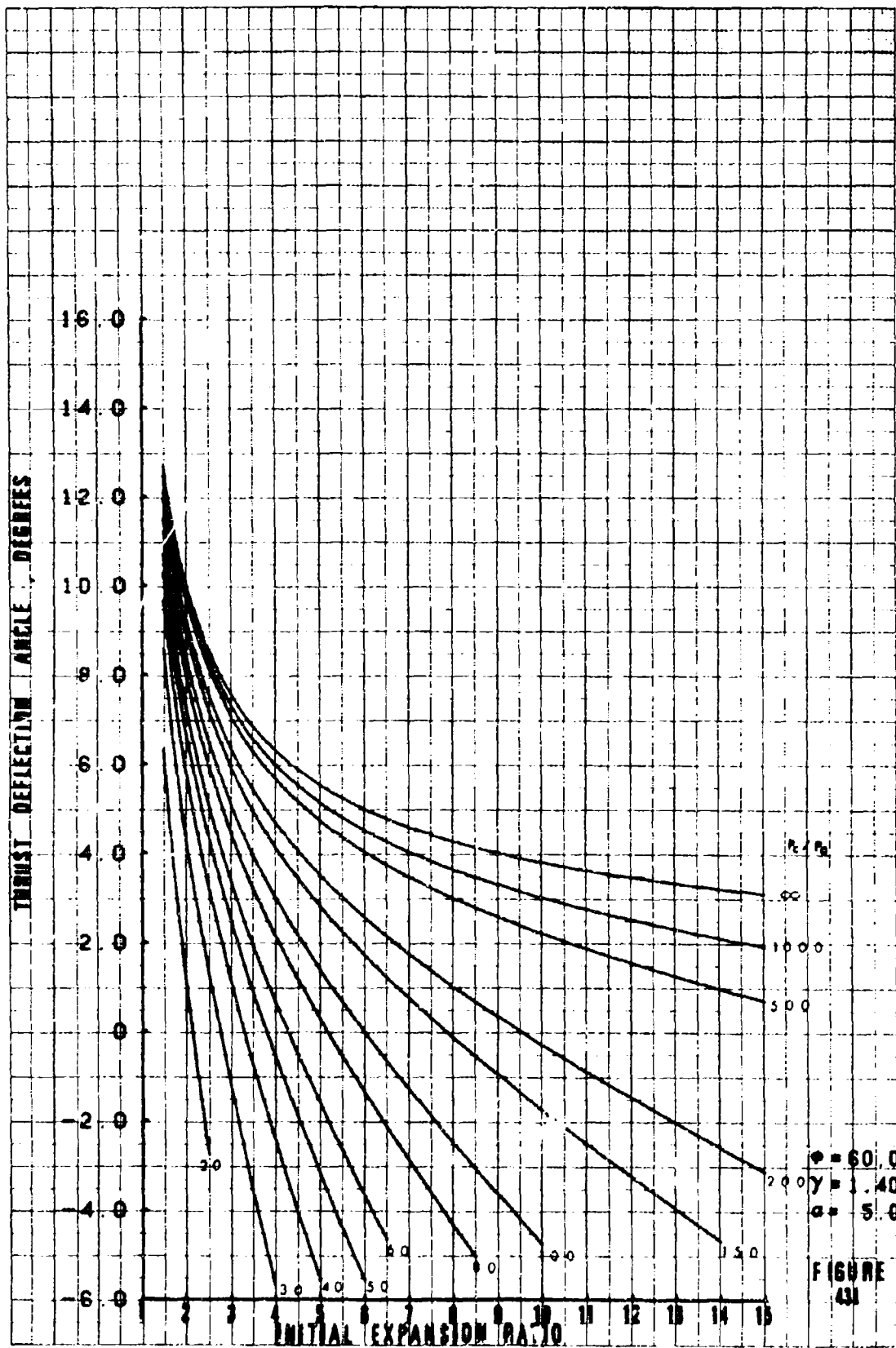
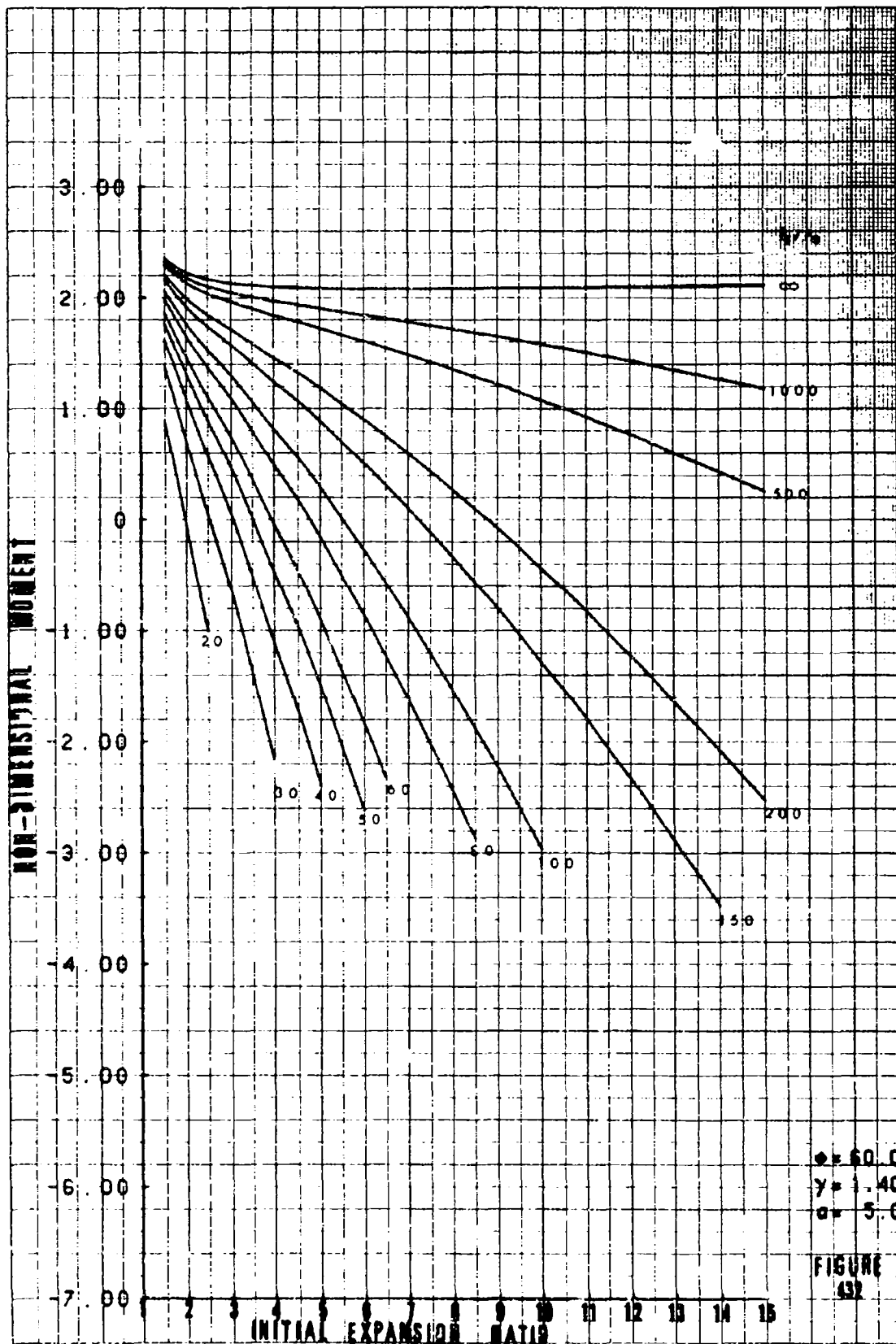
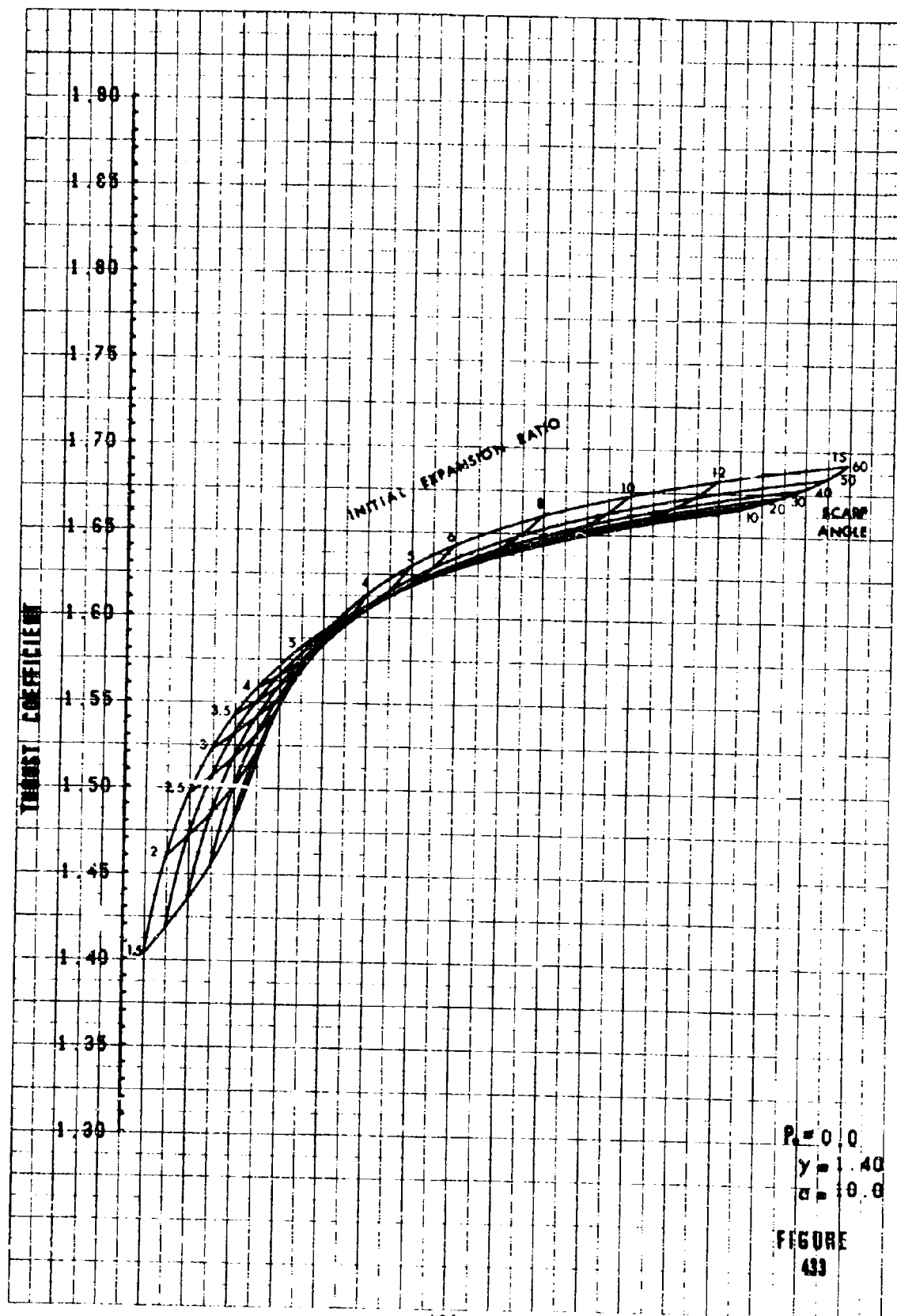
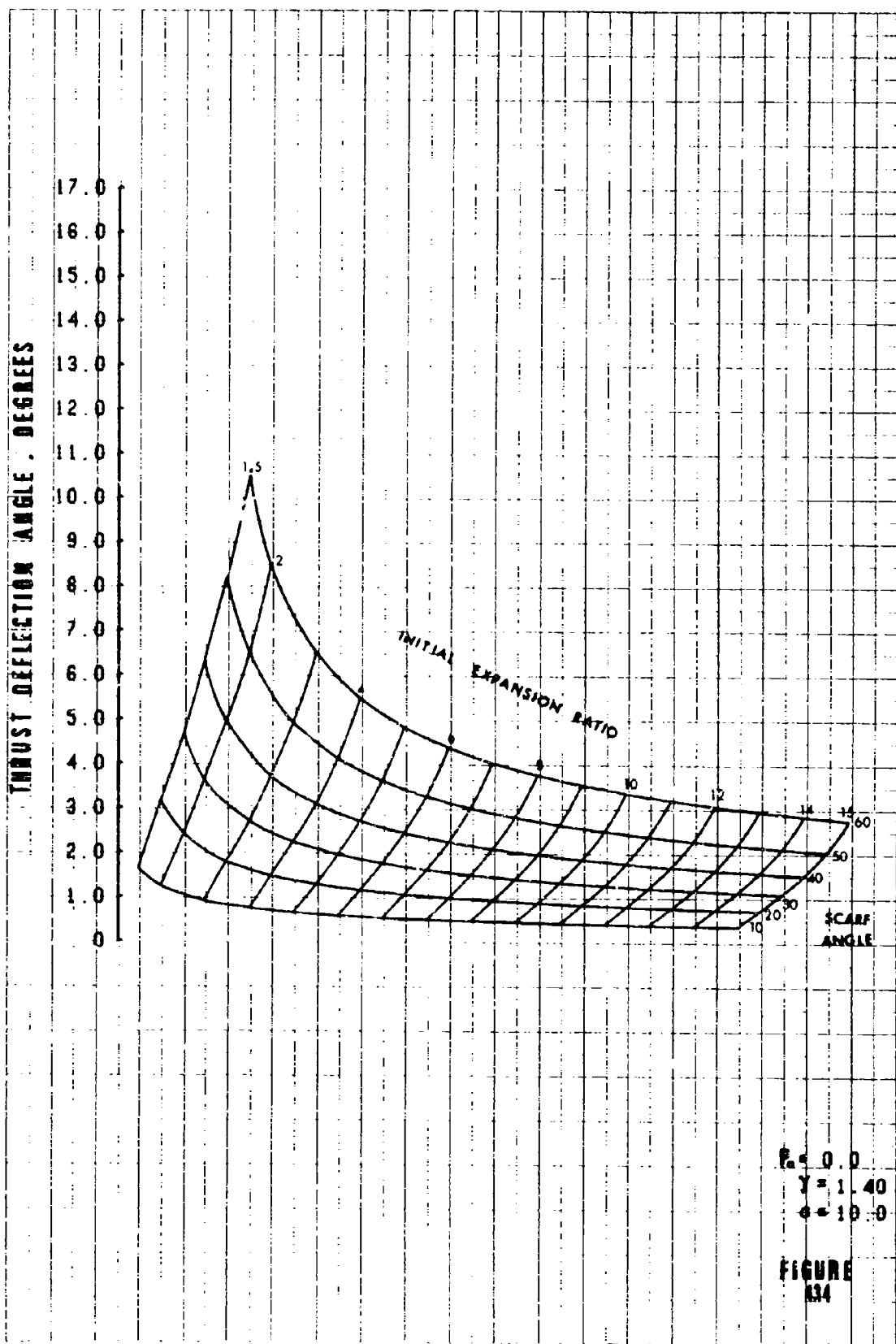
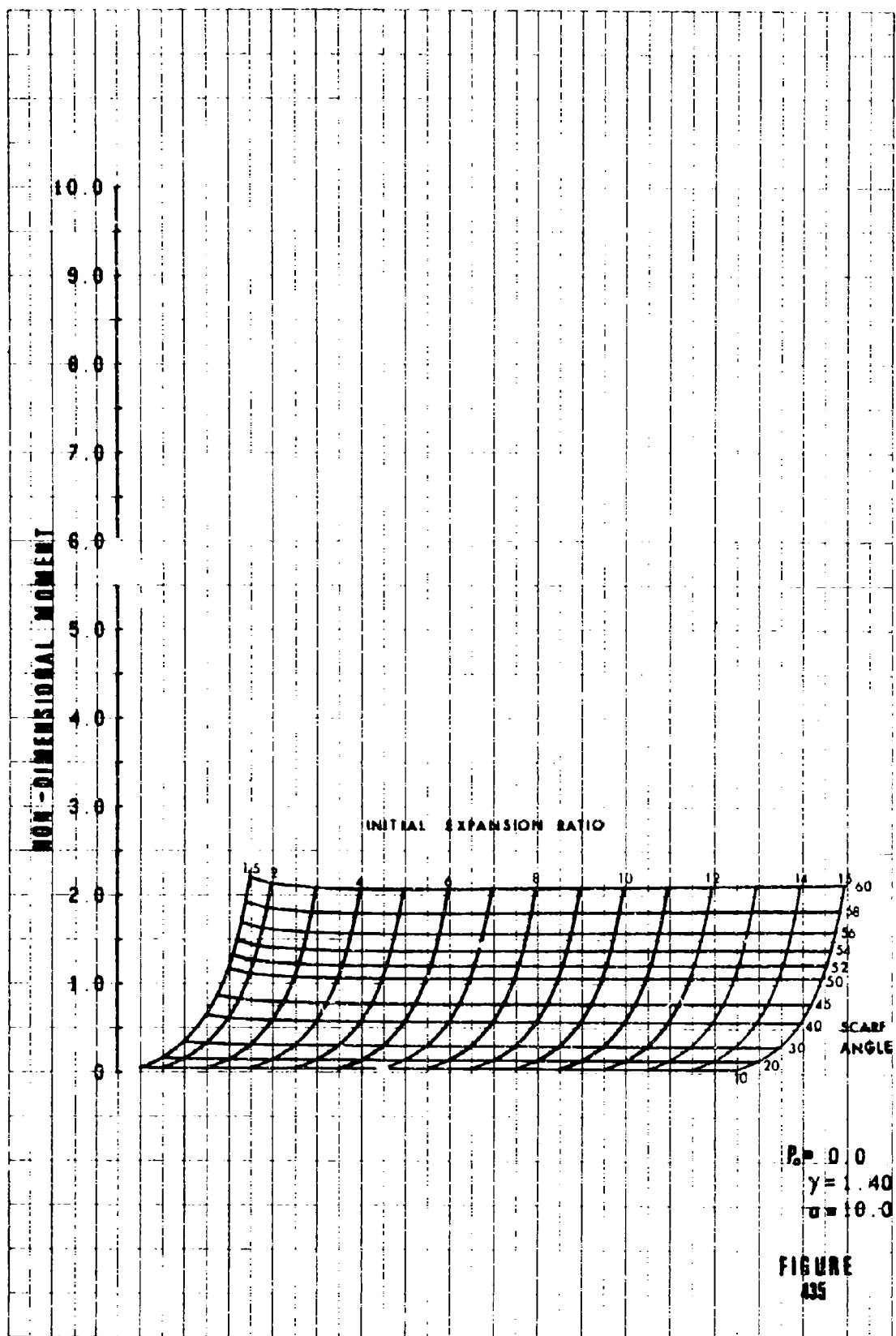


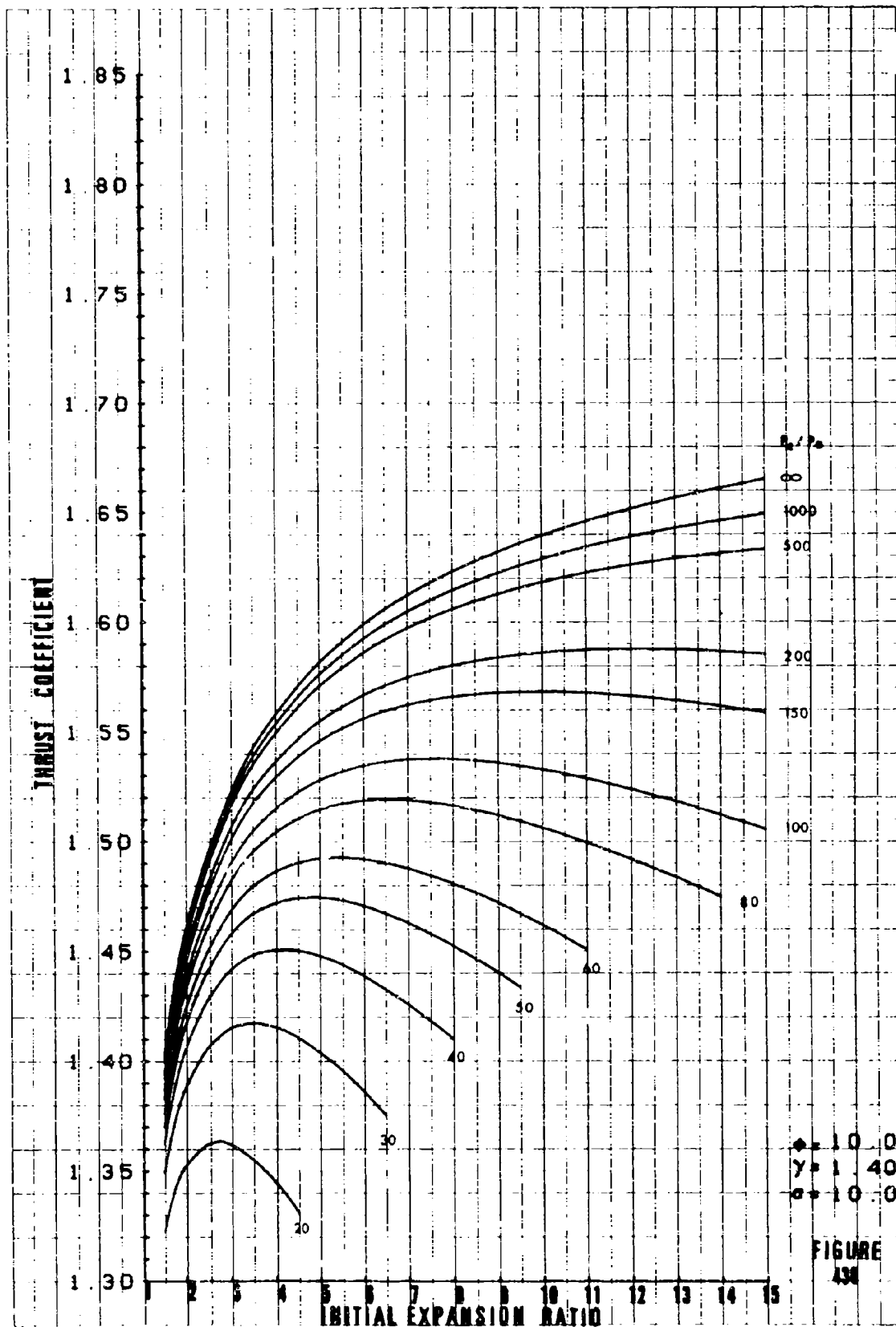
FIGURE 431

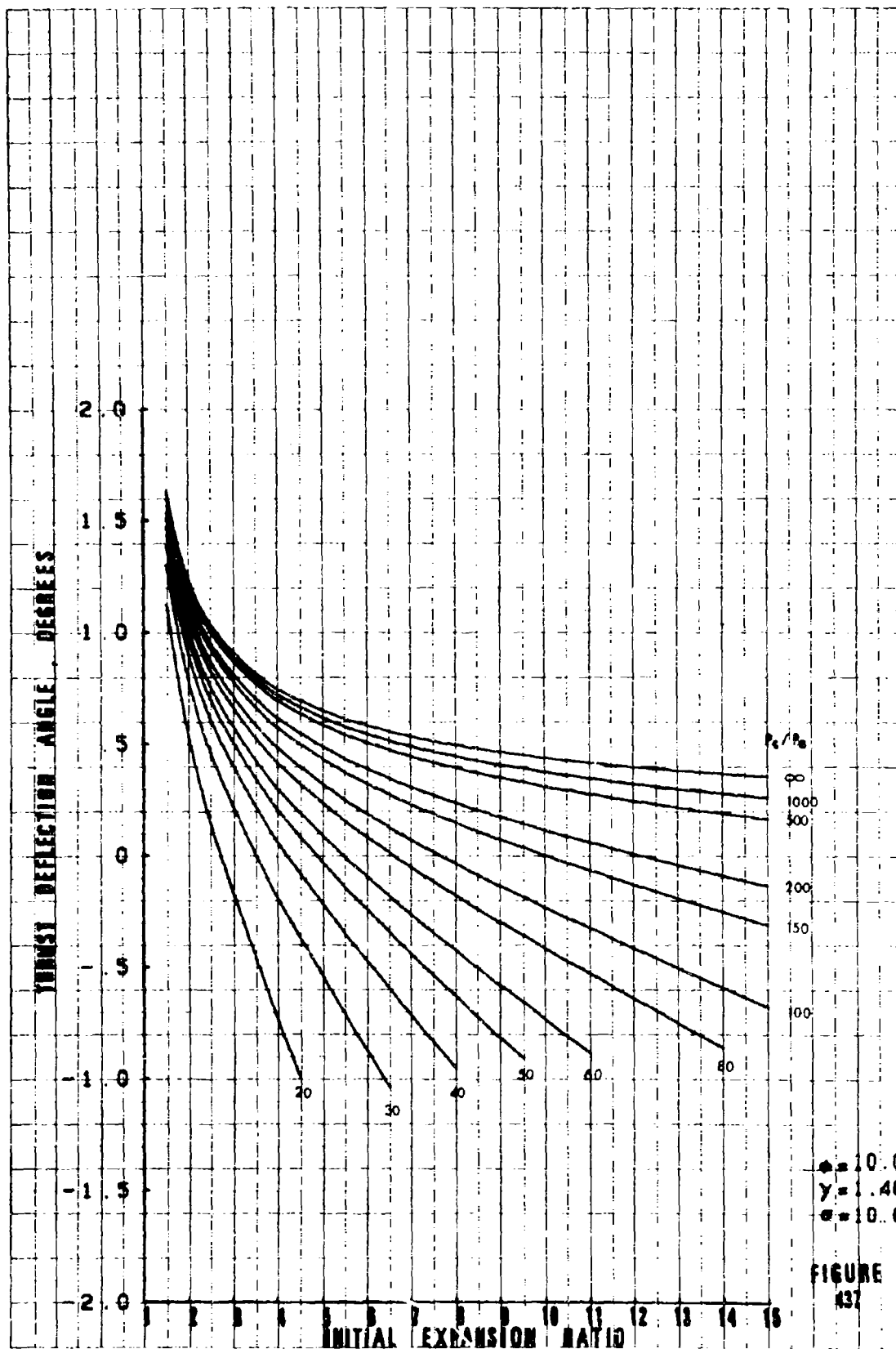


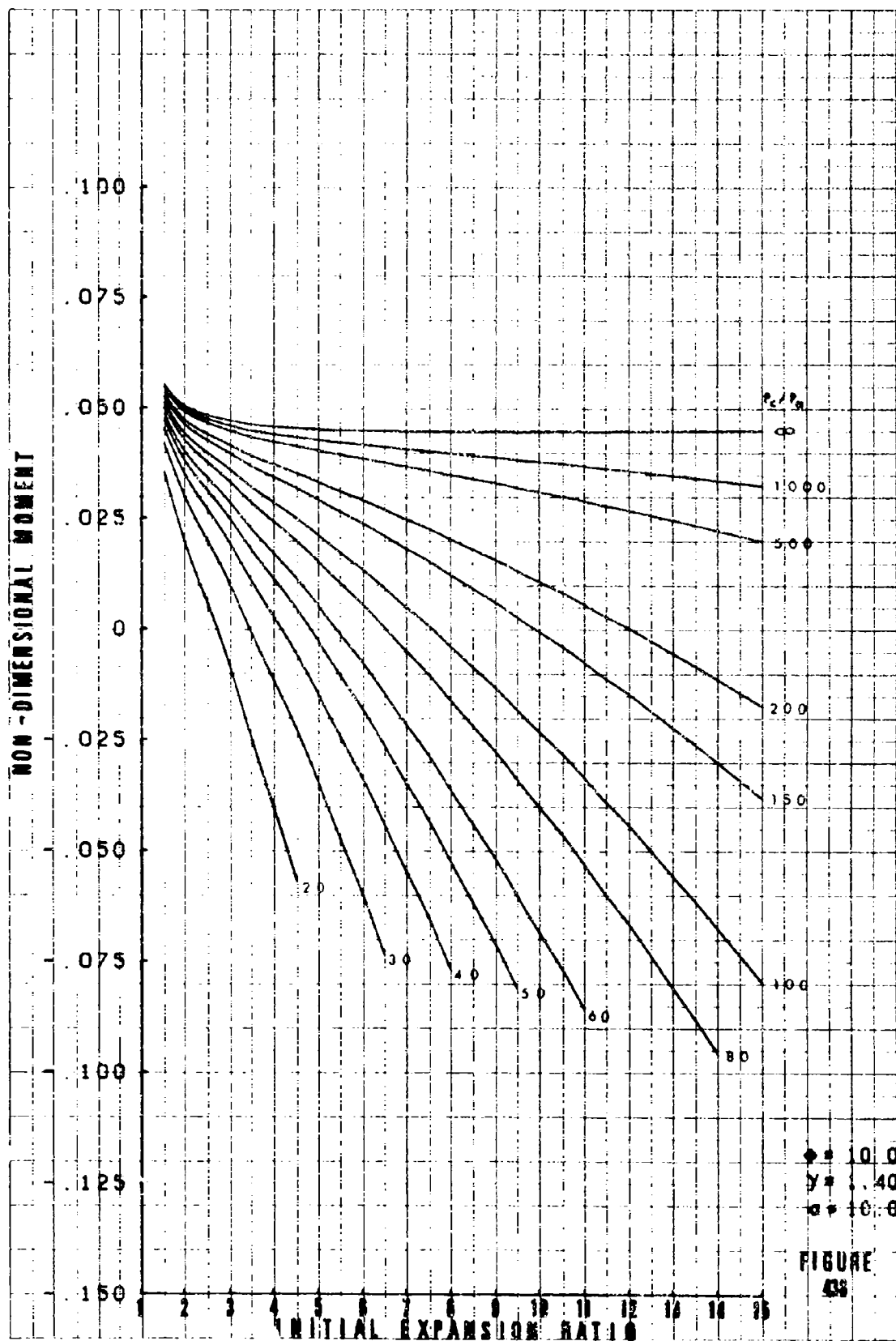












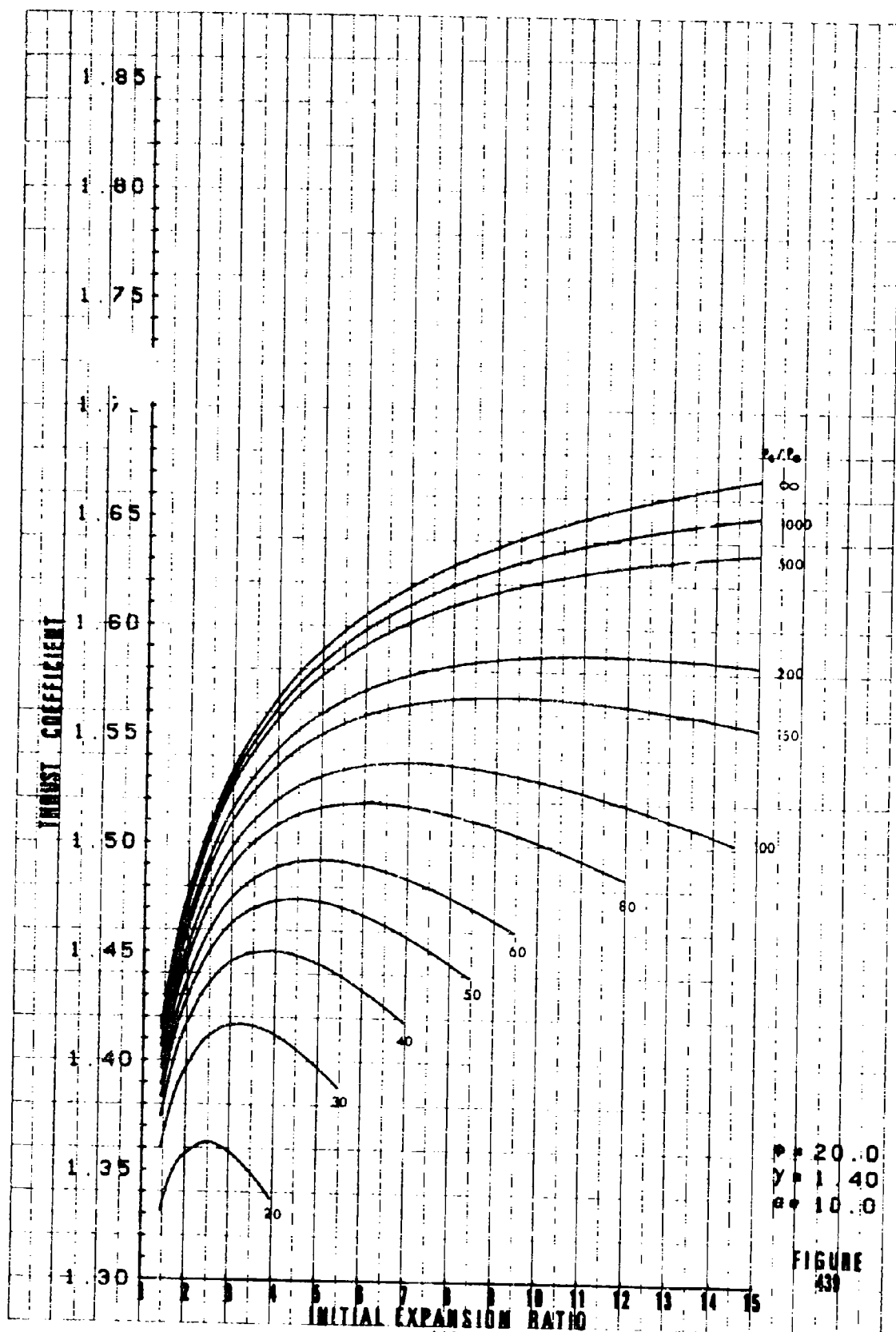
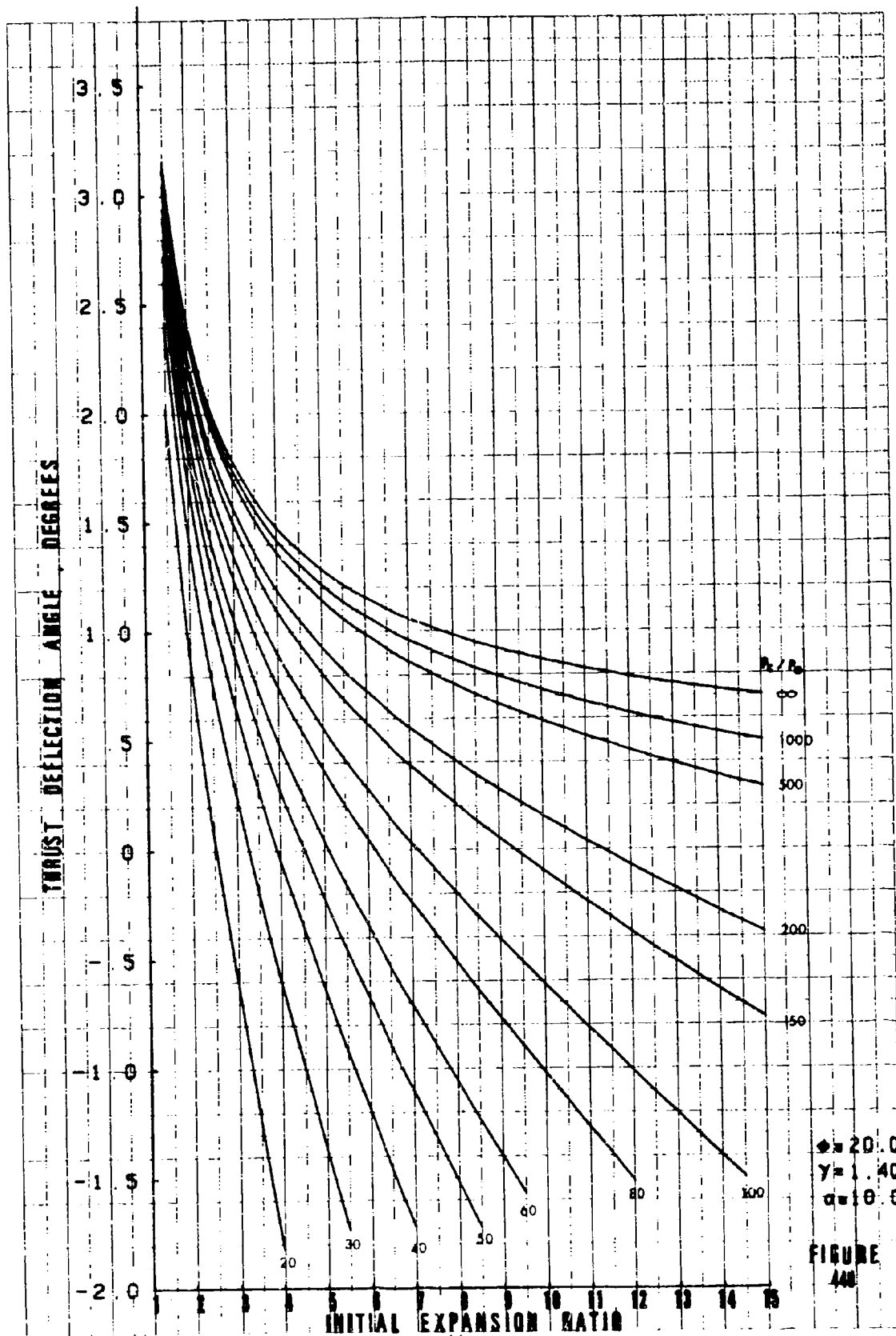
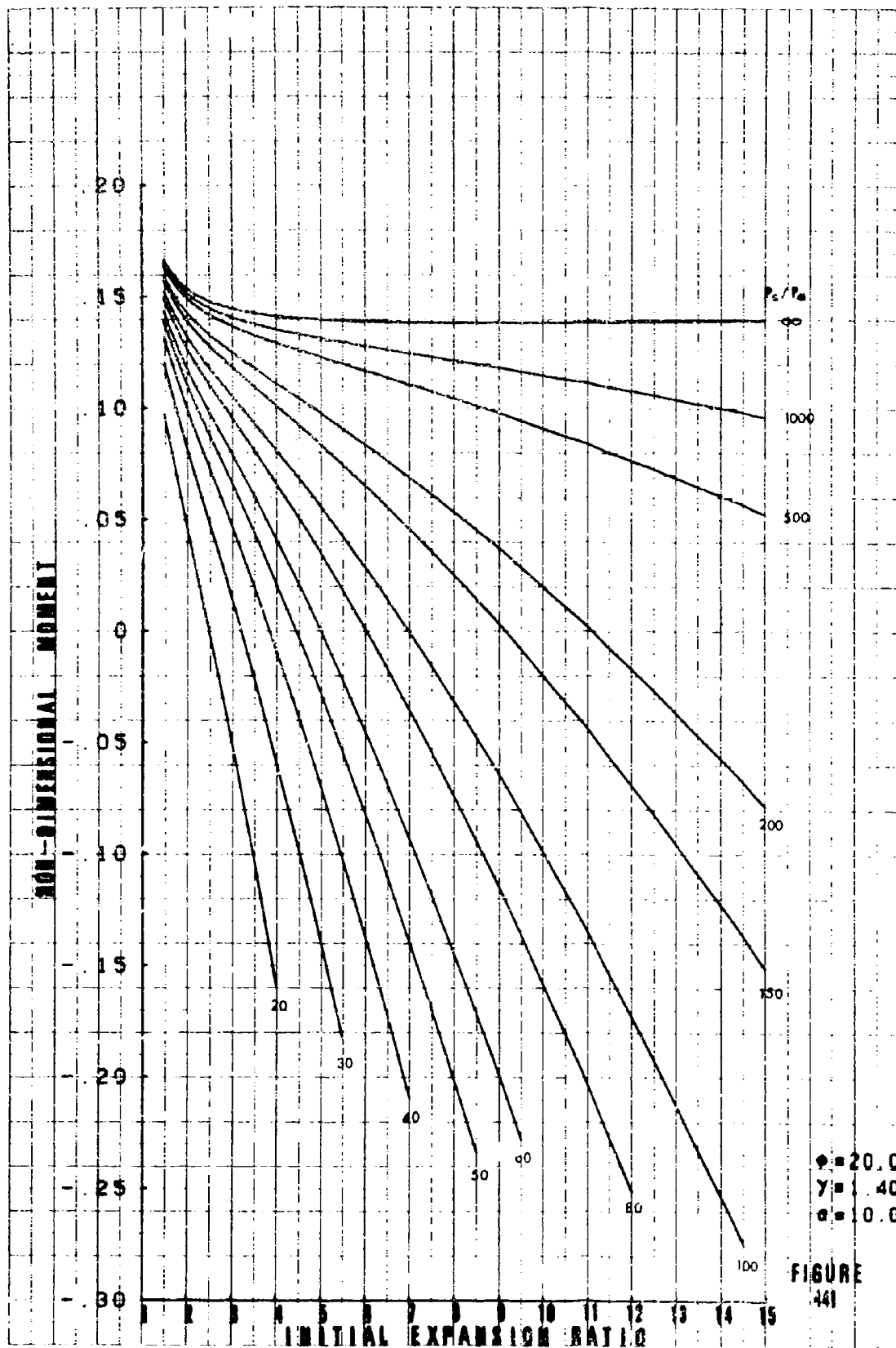
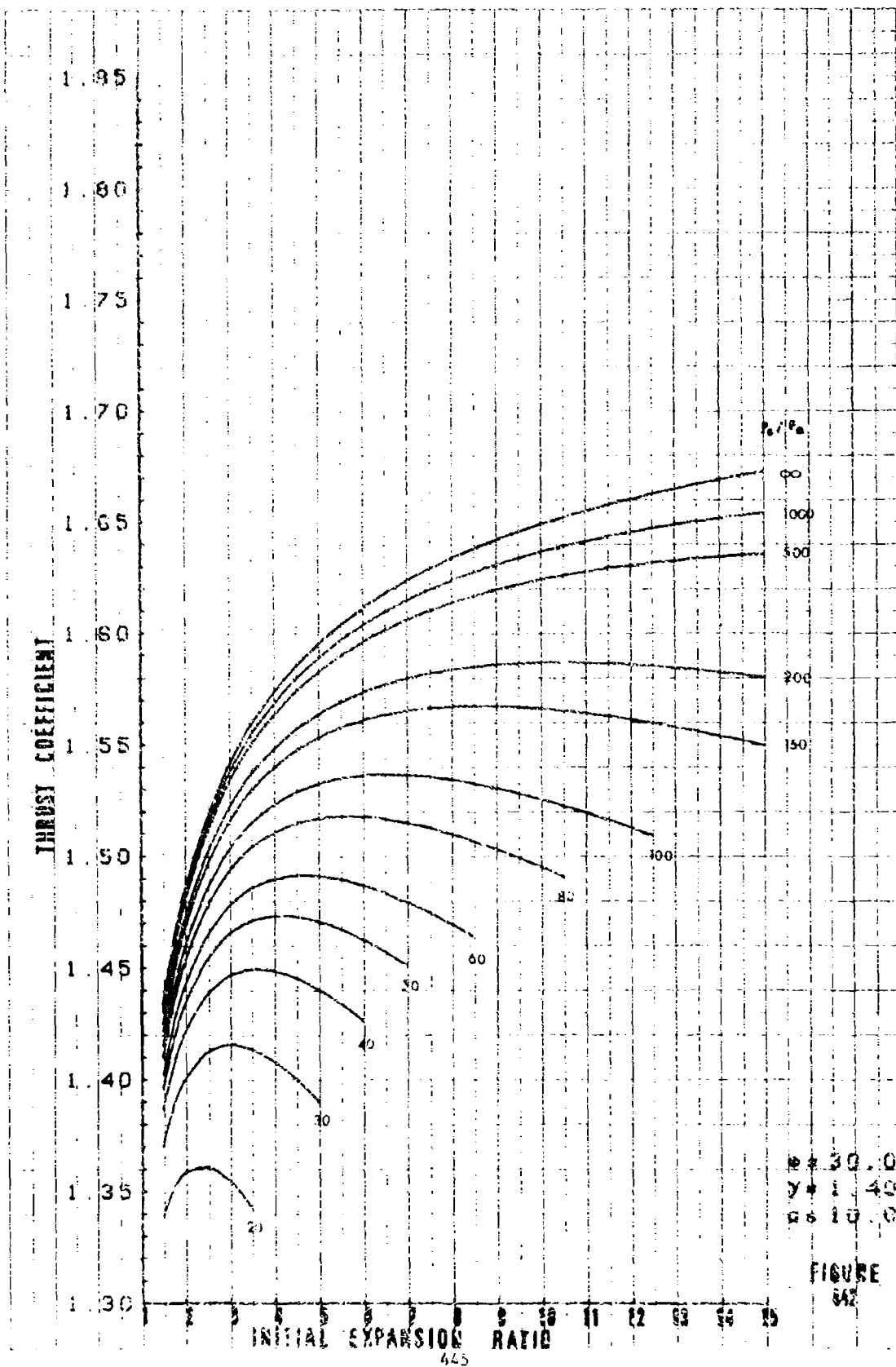
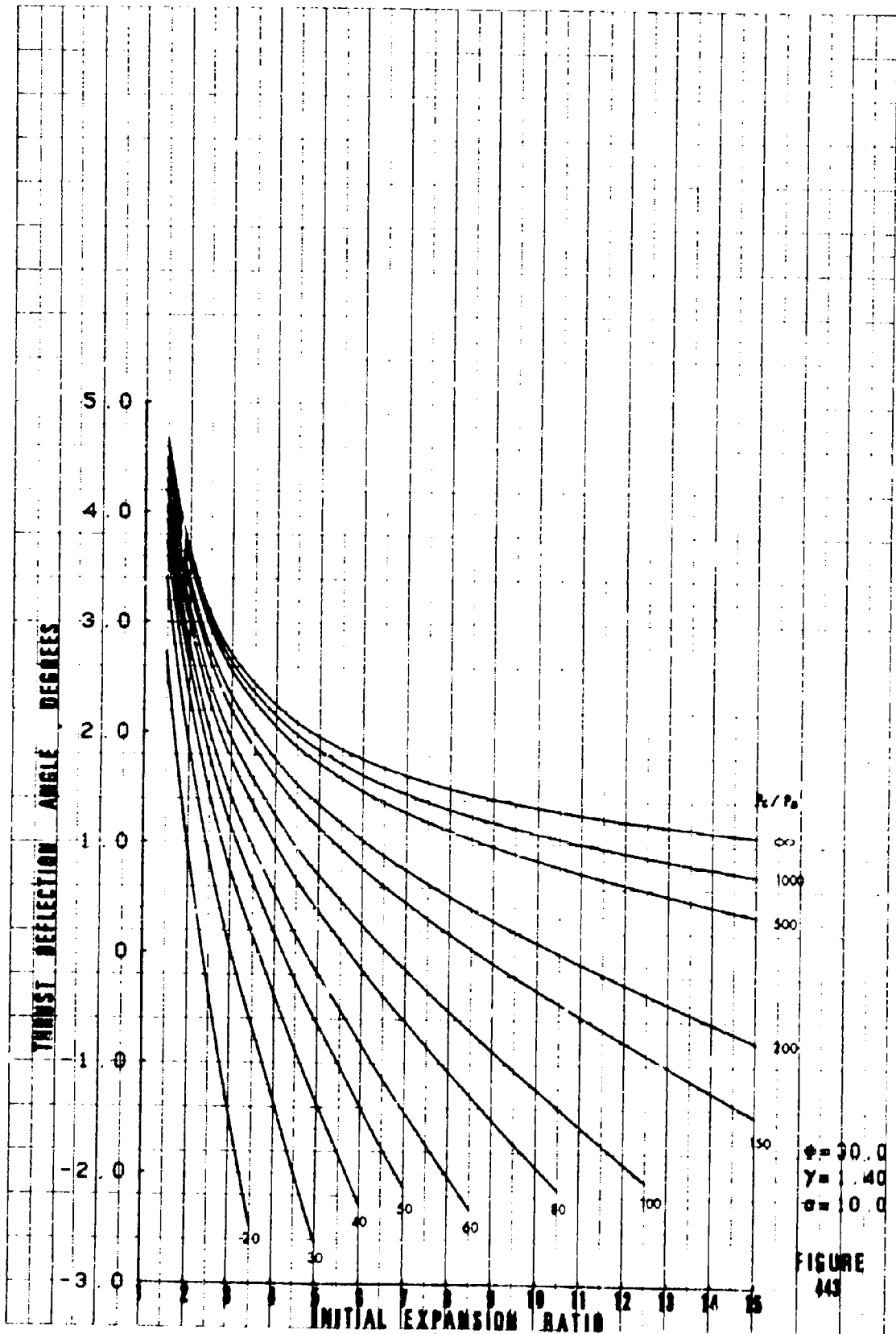


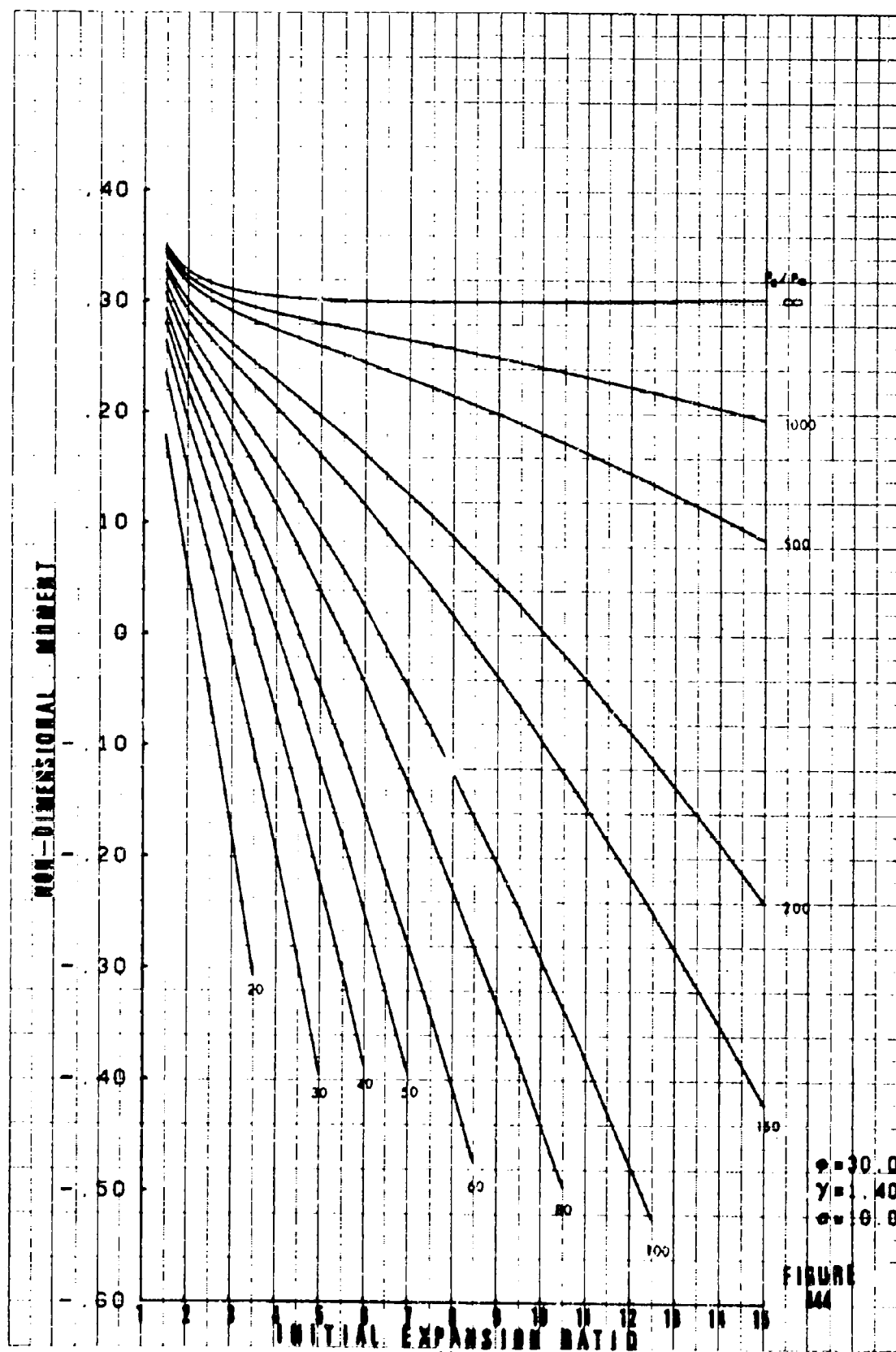
FIGURE 439

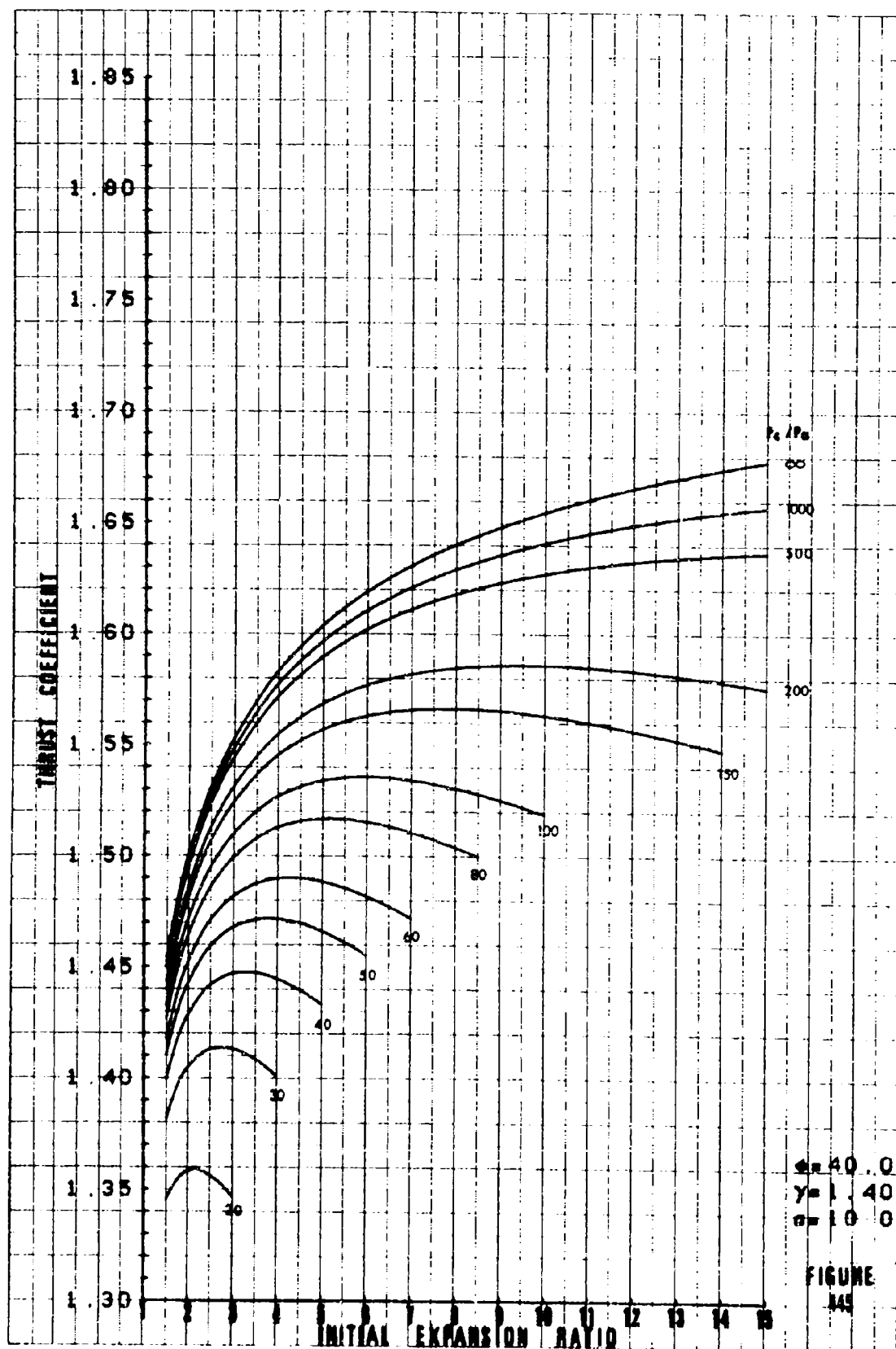


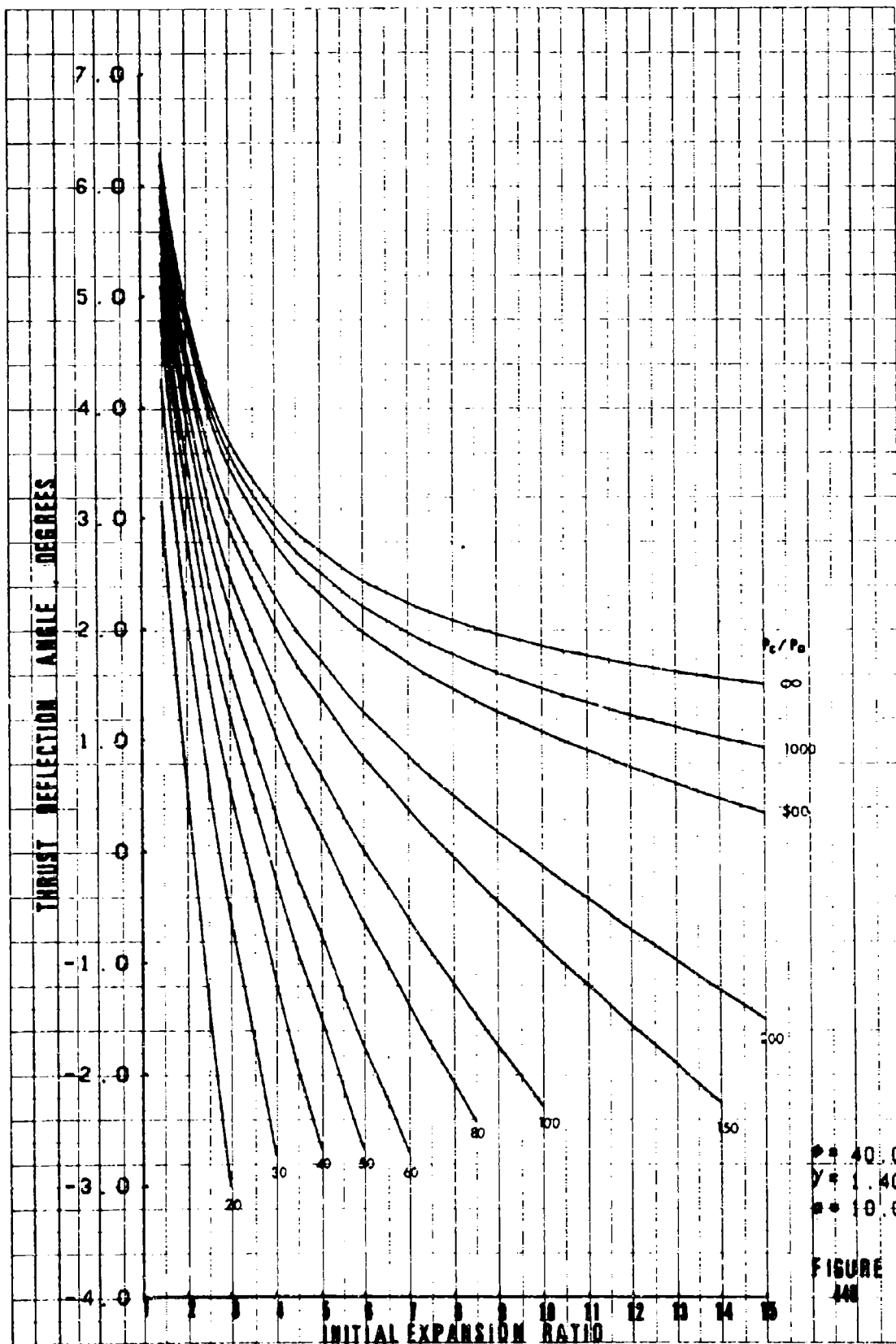


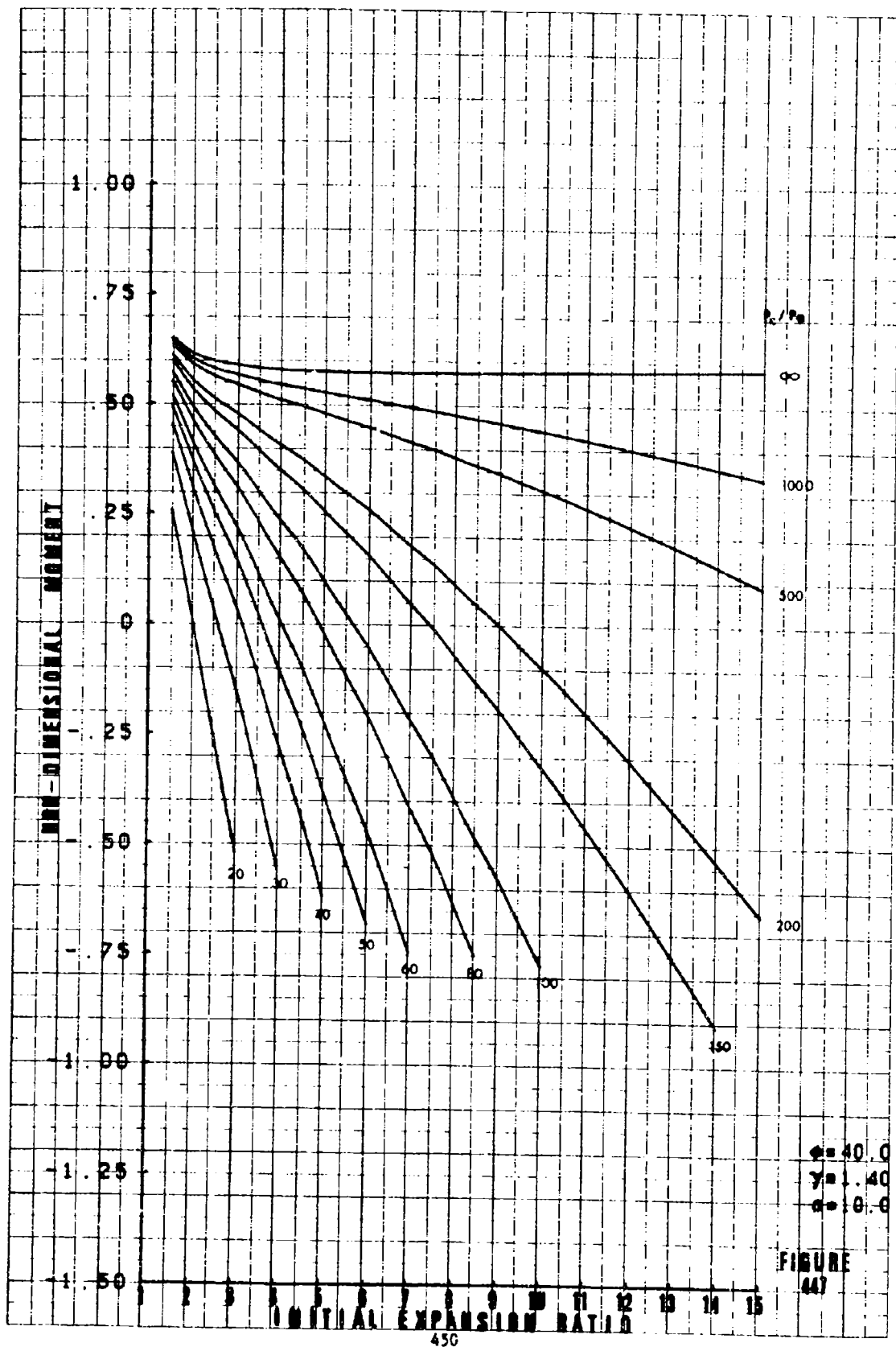












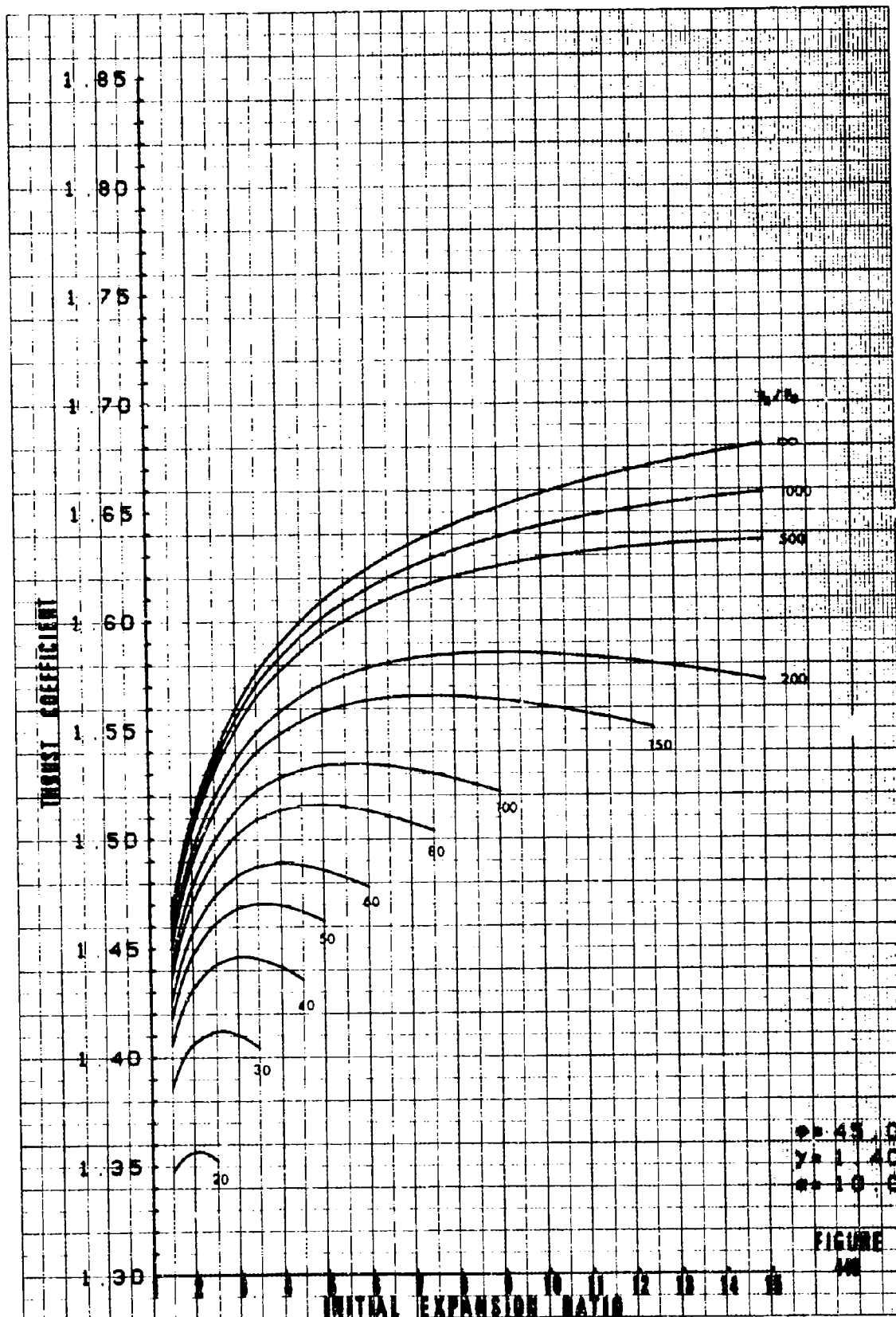


FIGURE
10

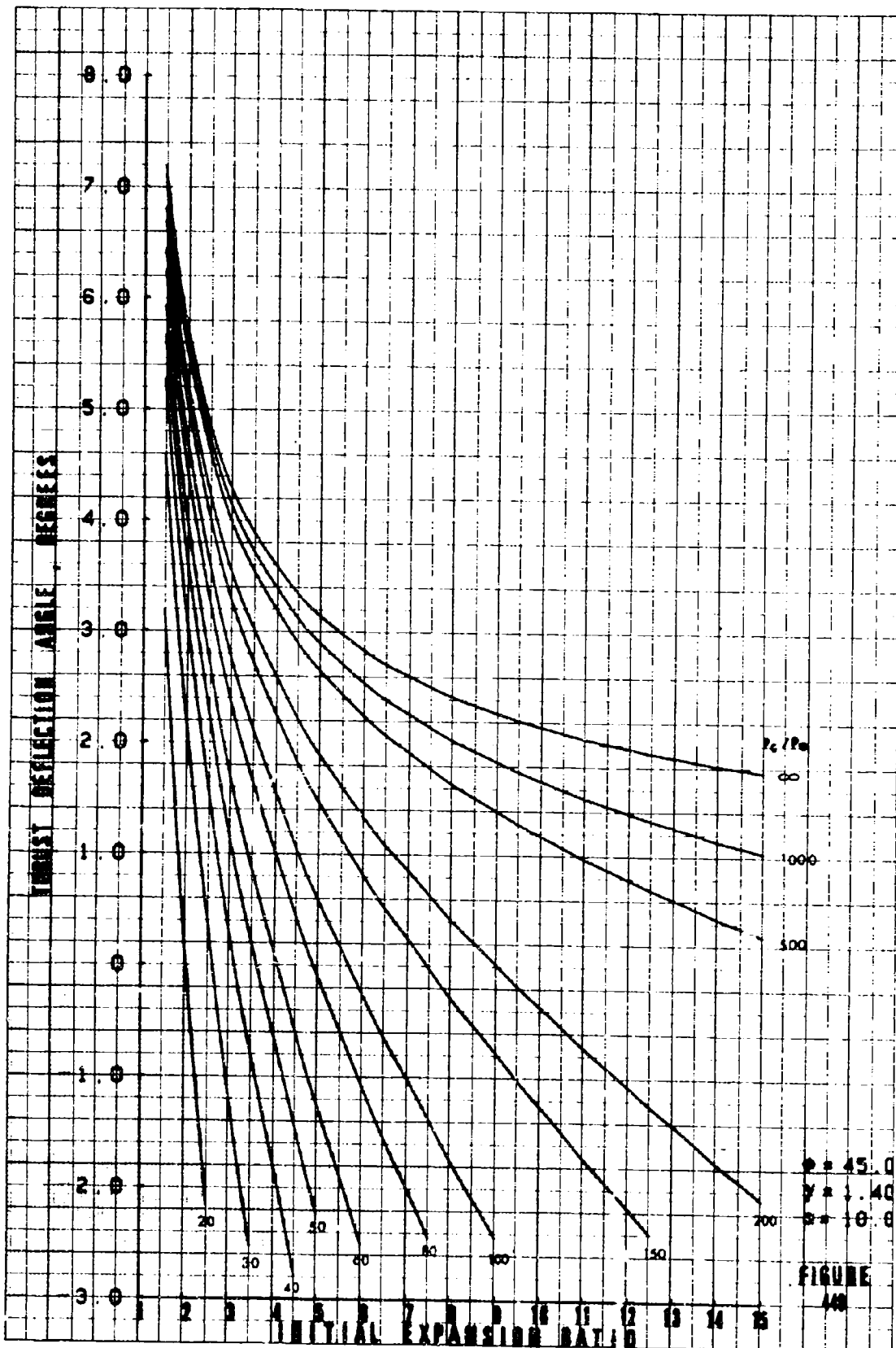
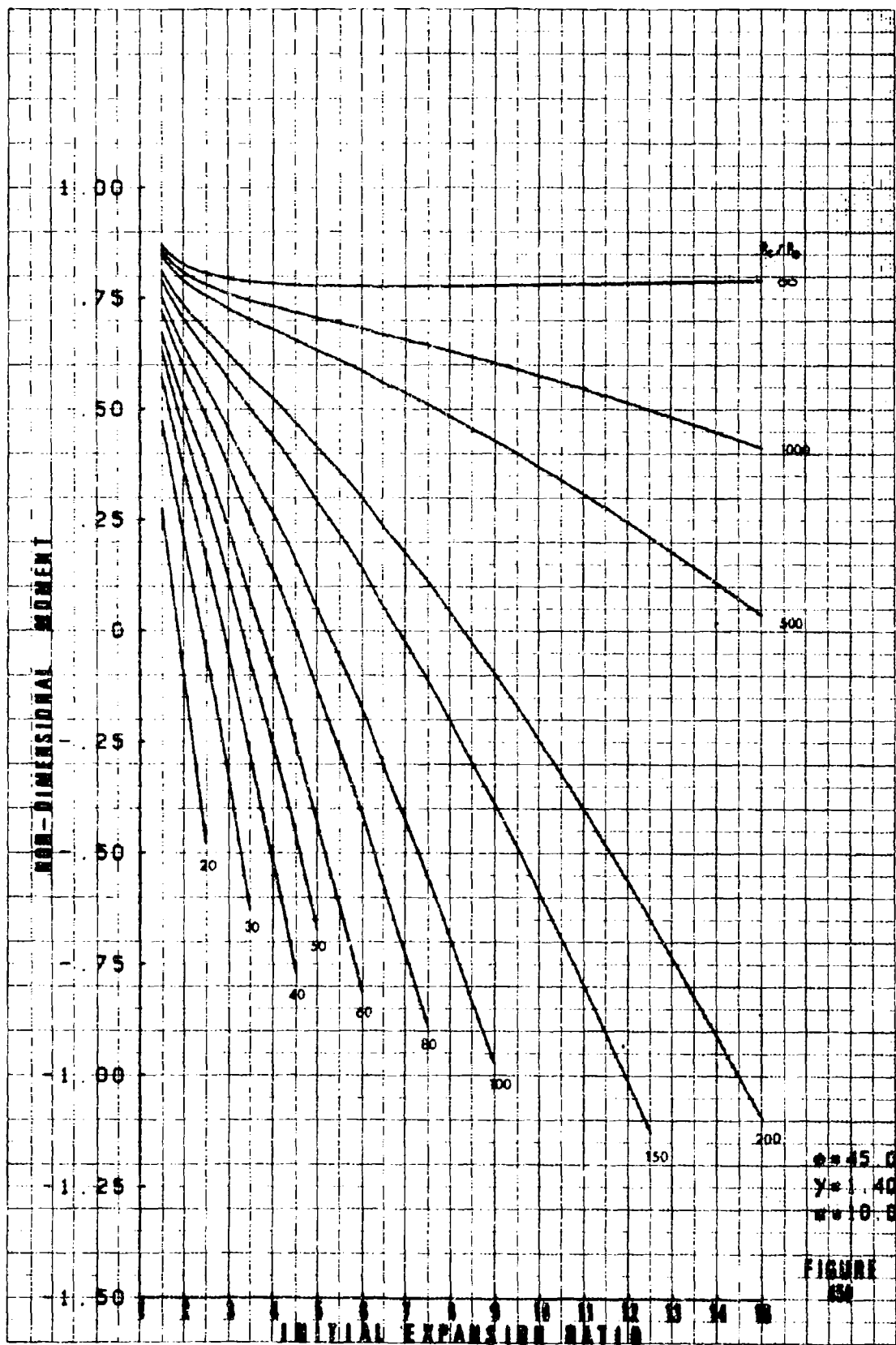
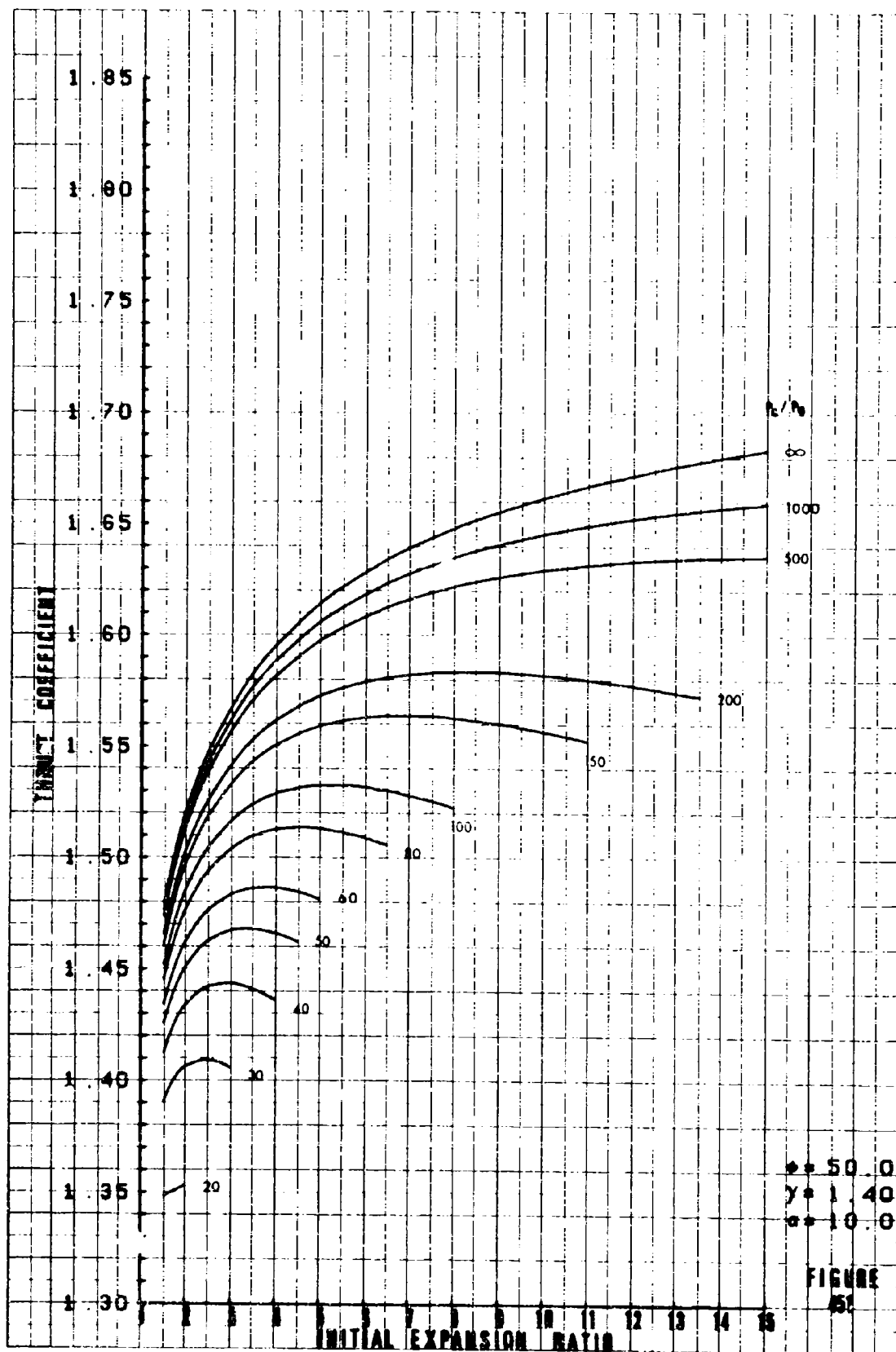


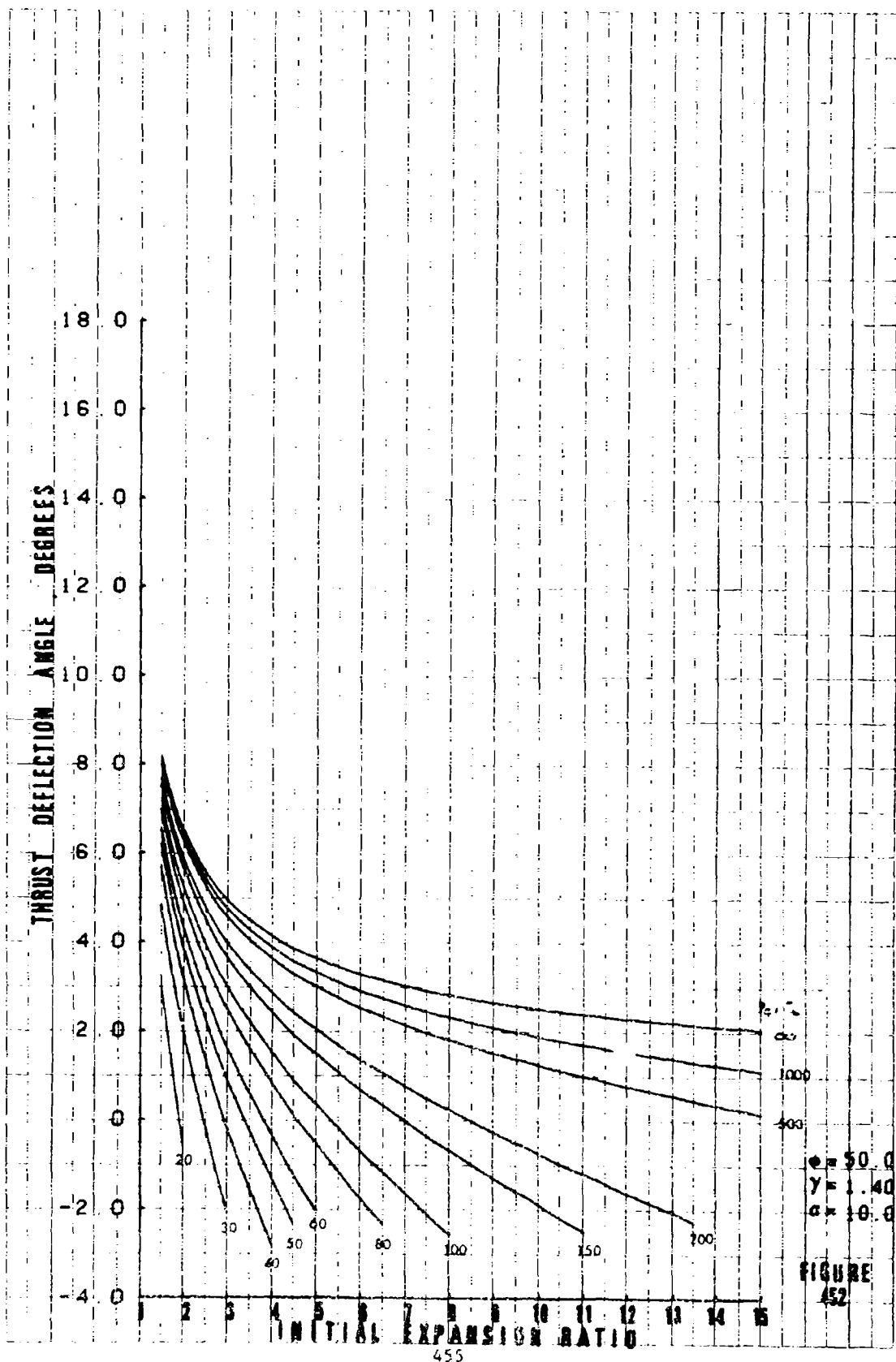
FIGURE 400



$\phi = 45.0$
 $\gamma = 1.40$
 $\mu = 0.8$

FIGURE
150





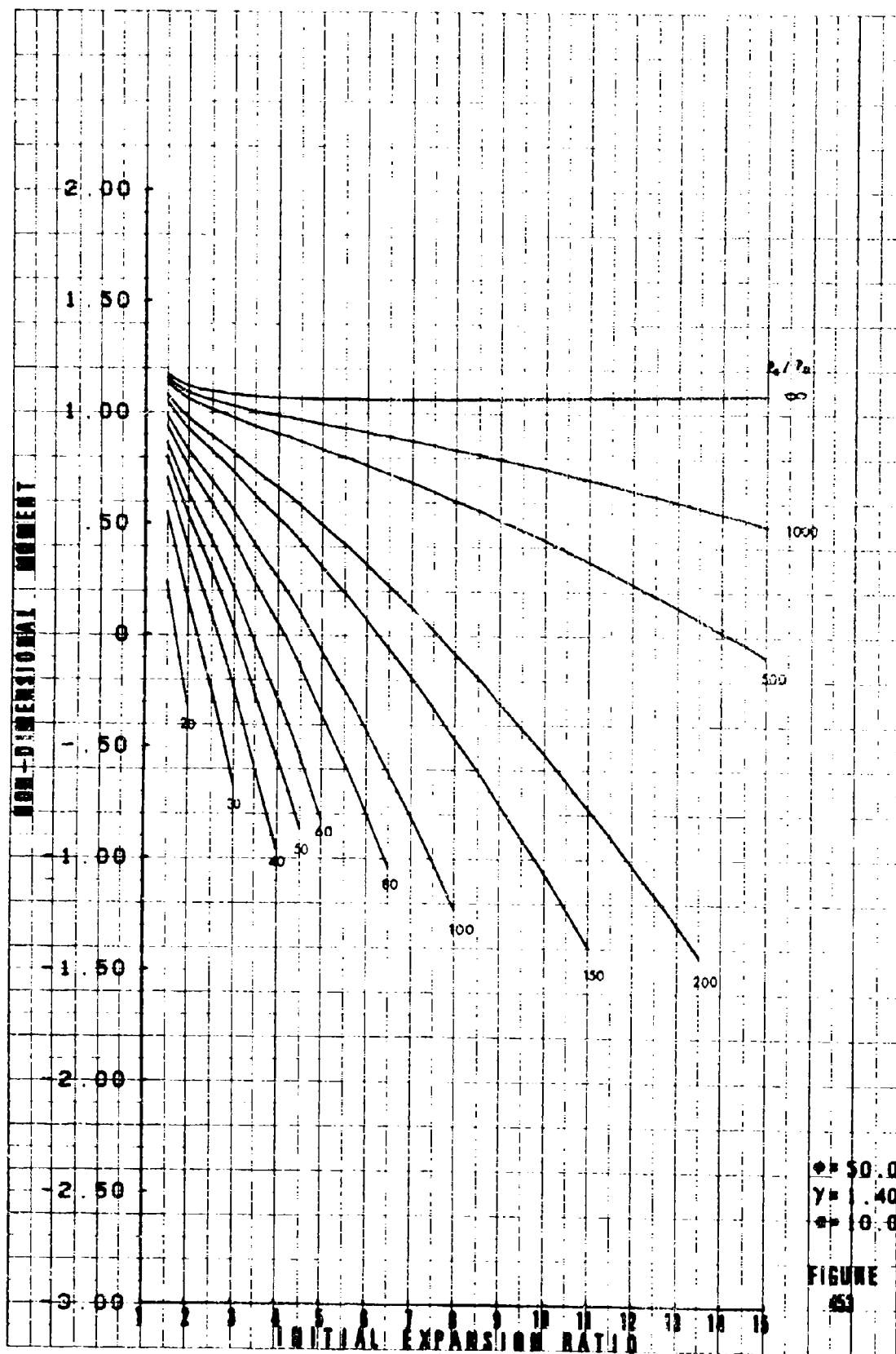
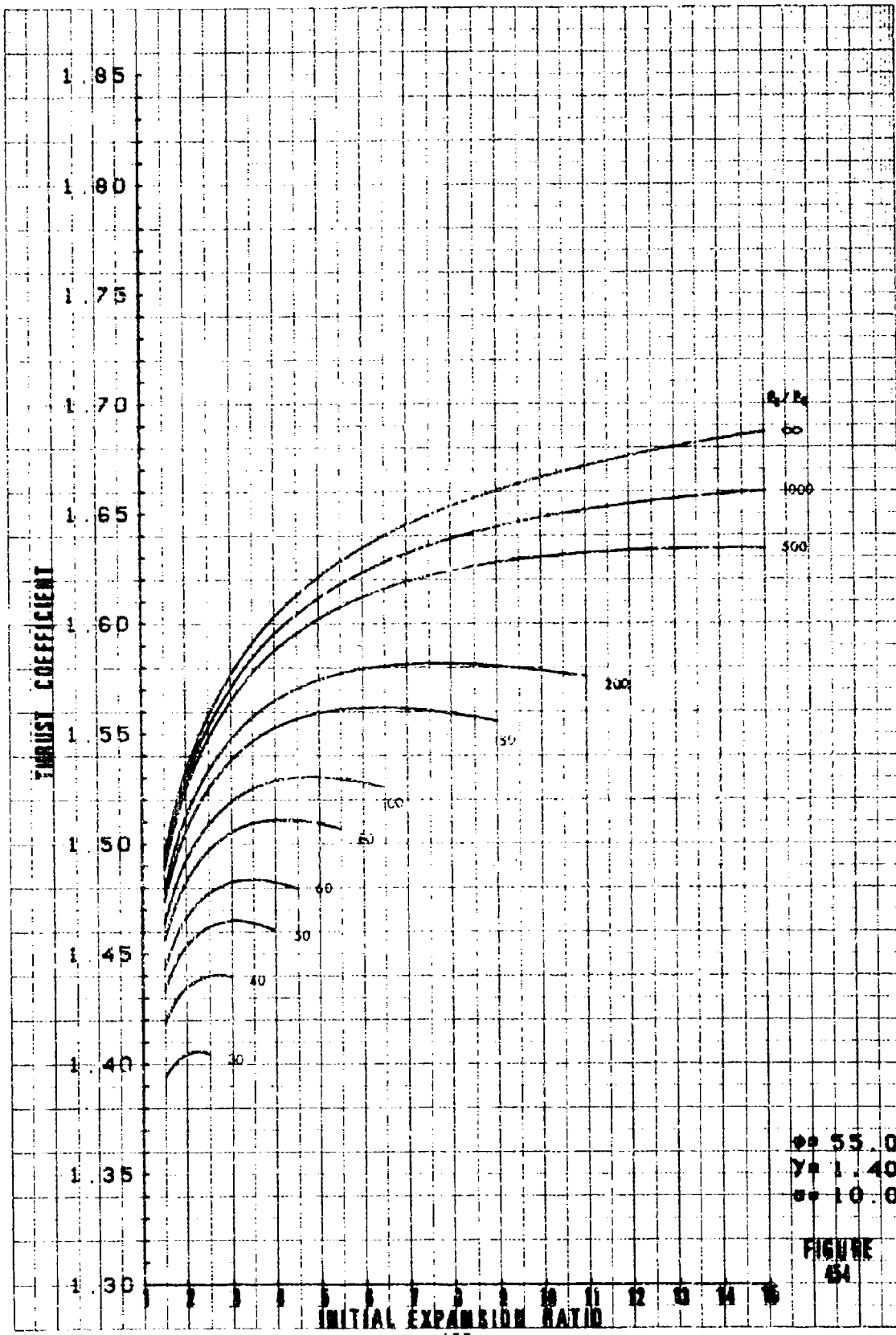


FIGURE 453



$\phi = 55.0$
 $\gamma = 1.40$
 $\phi = 10.0$

FIGURE 54

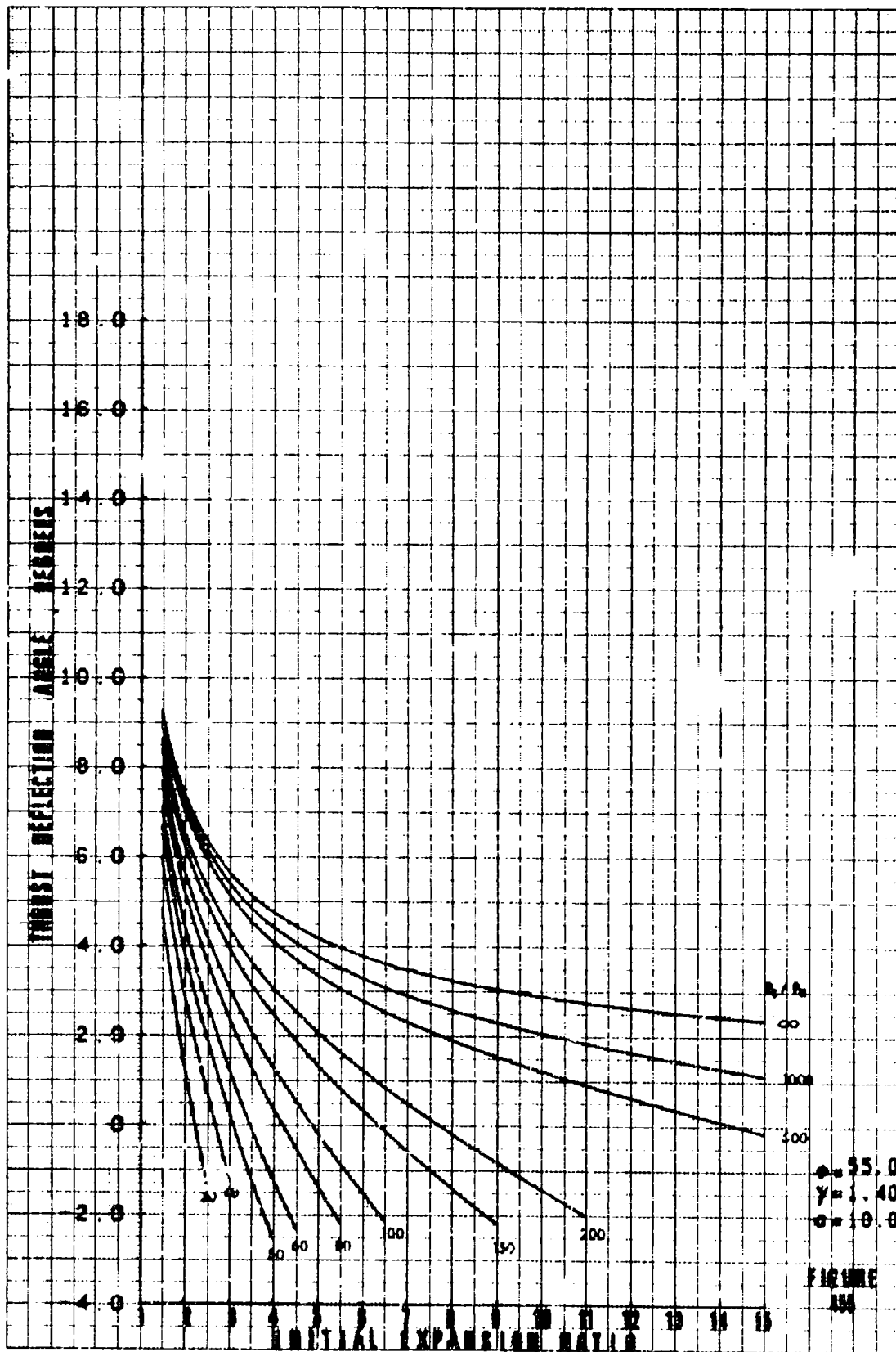
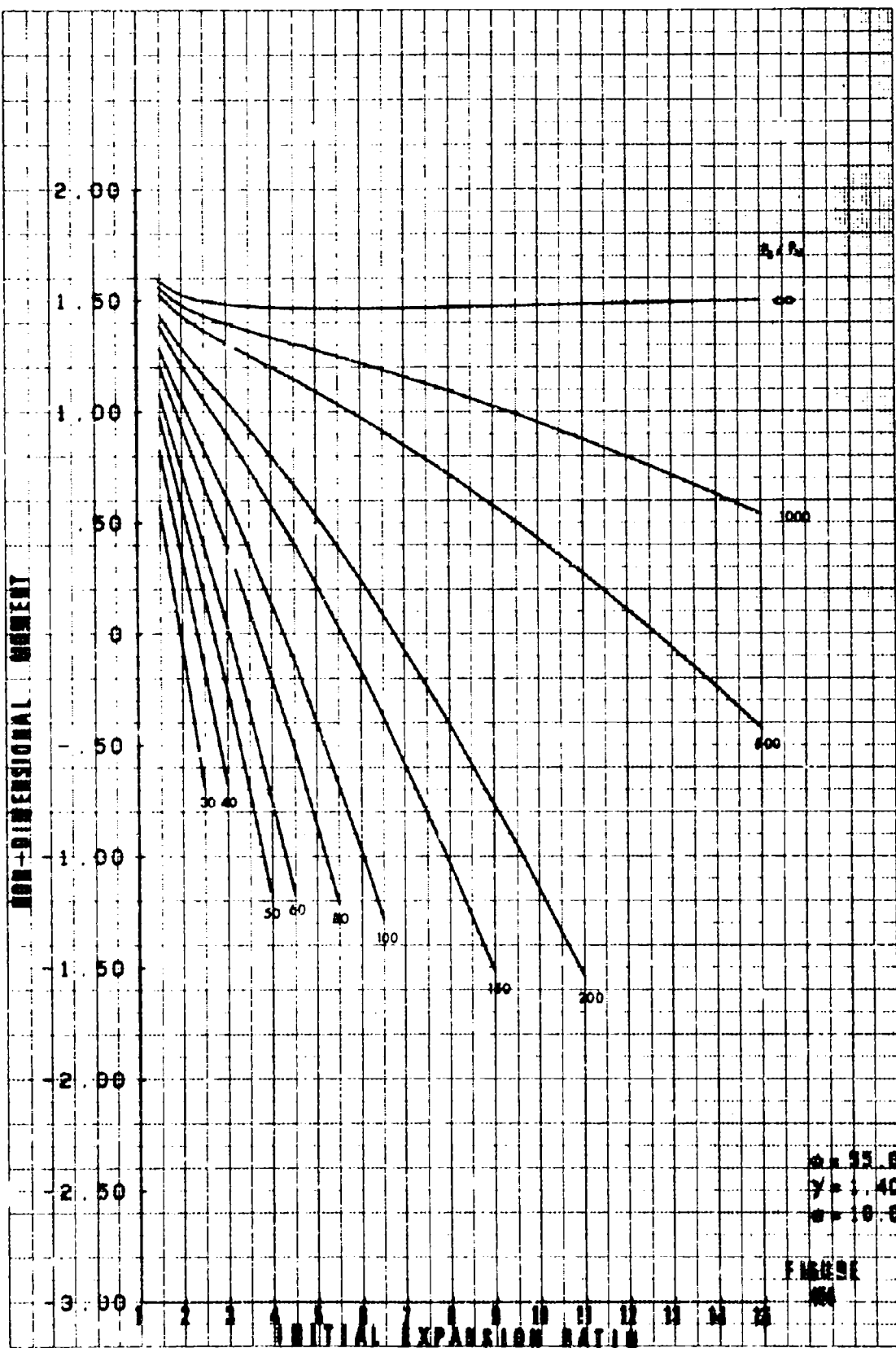
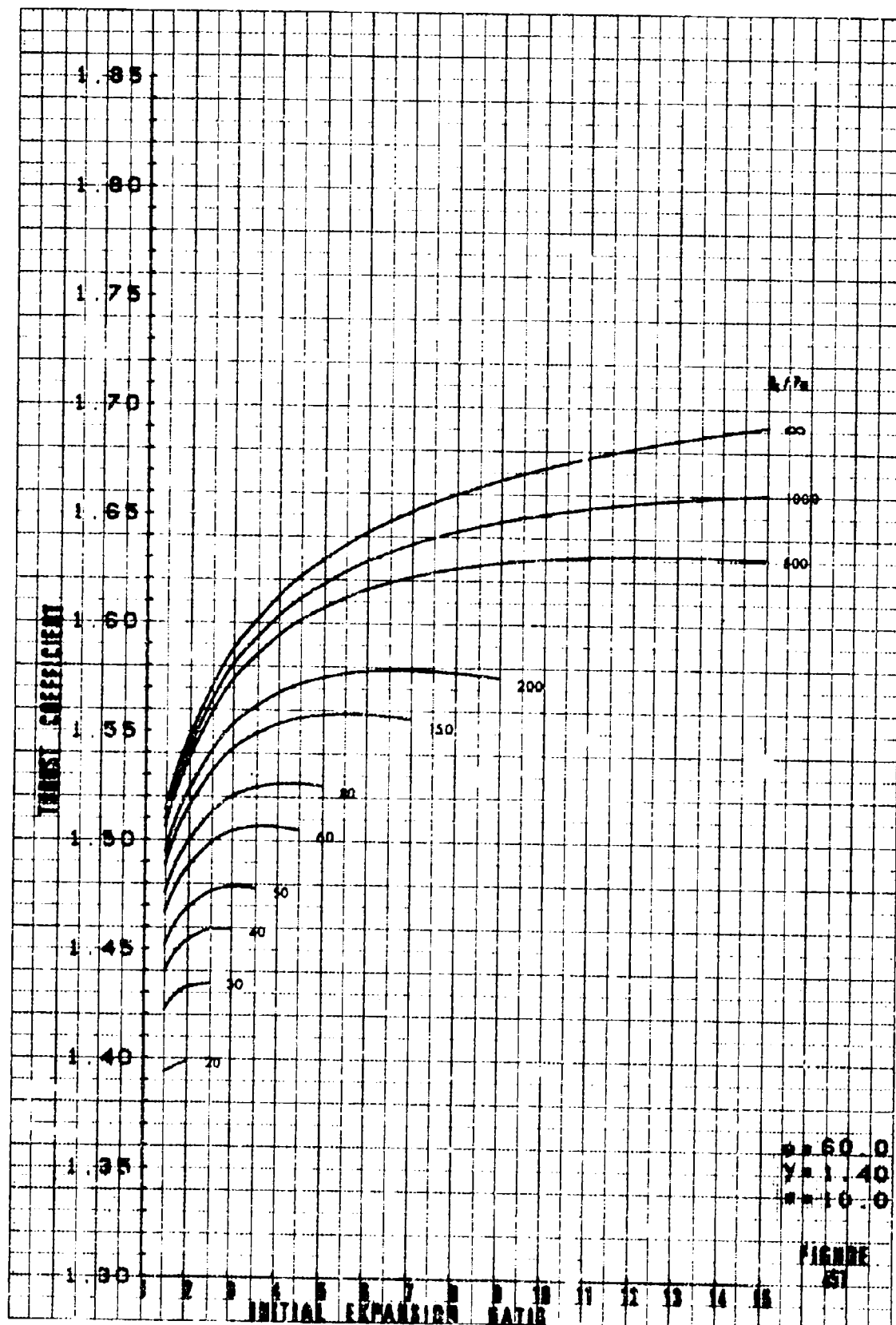
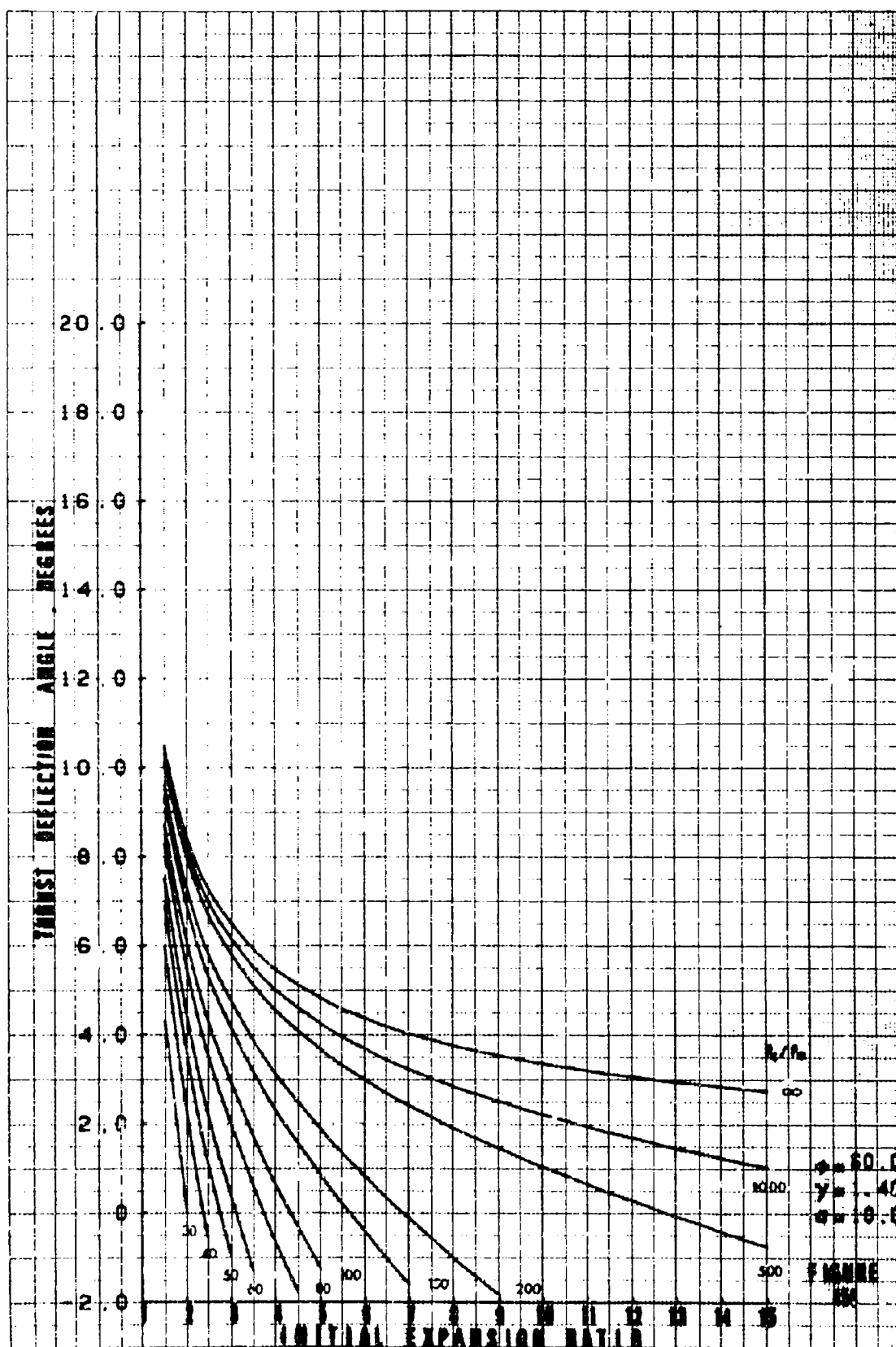
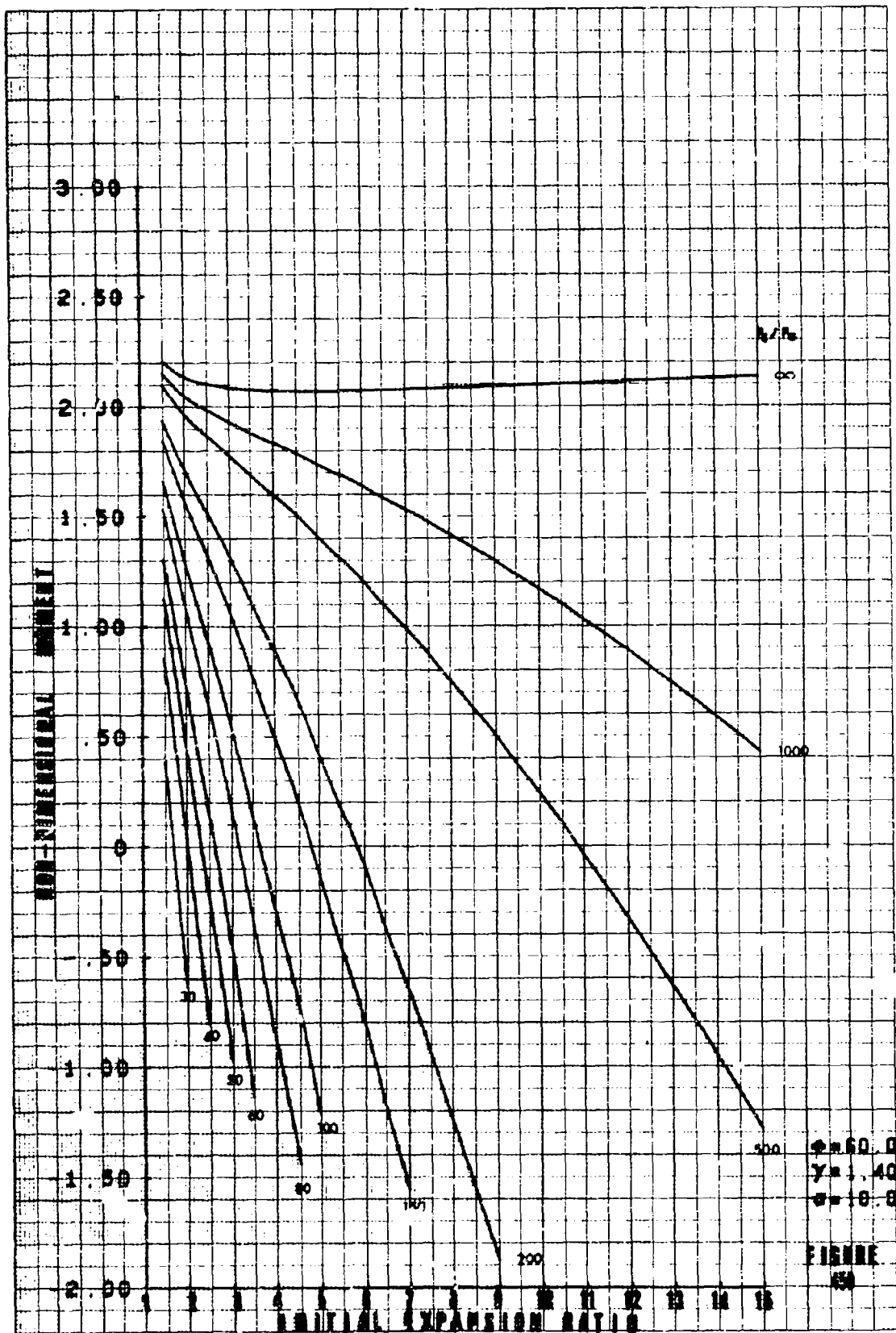


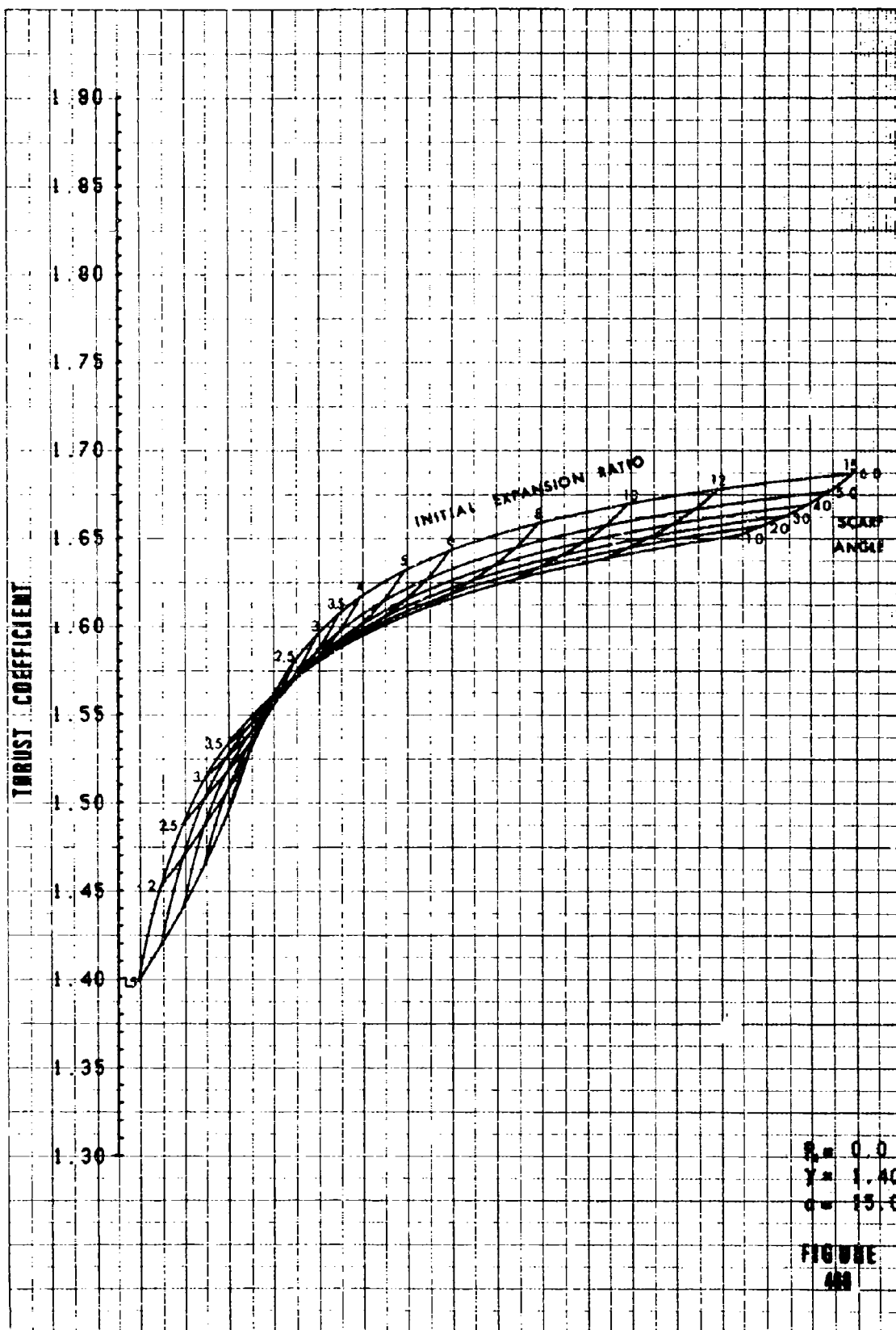
FIGURE 154

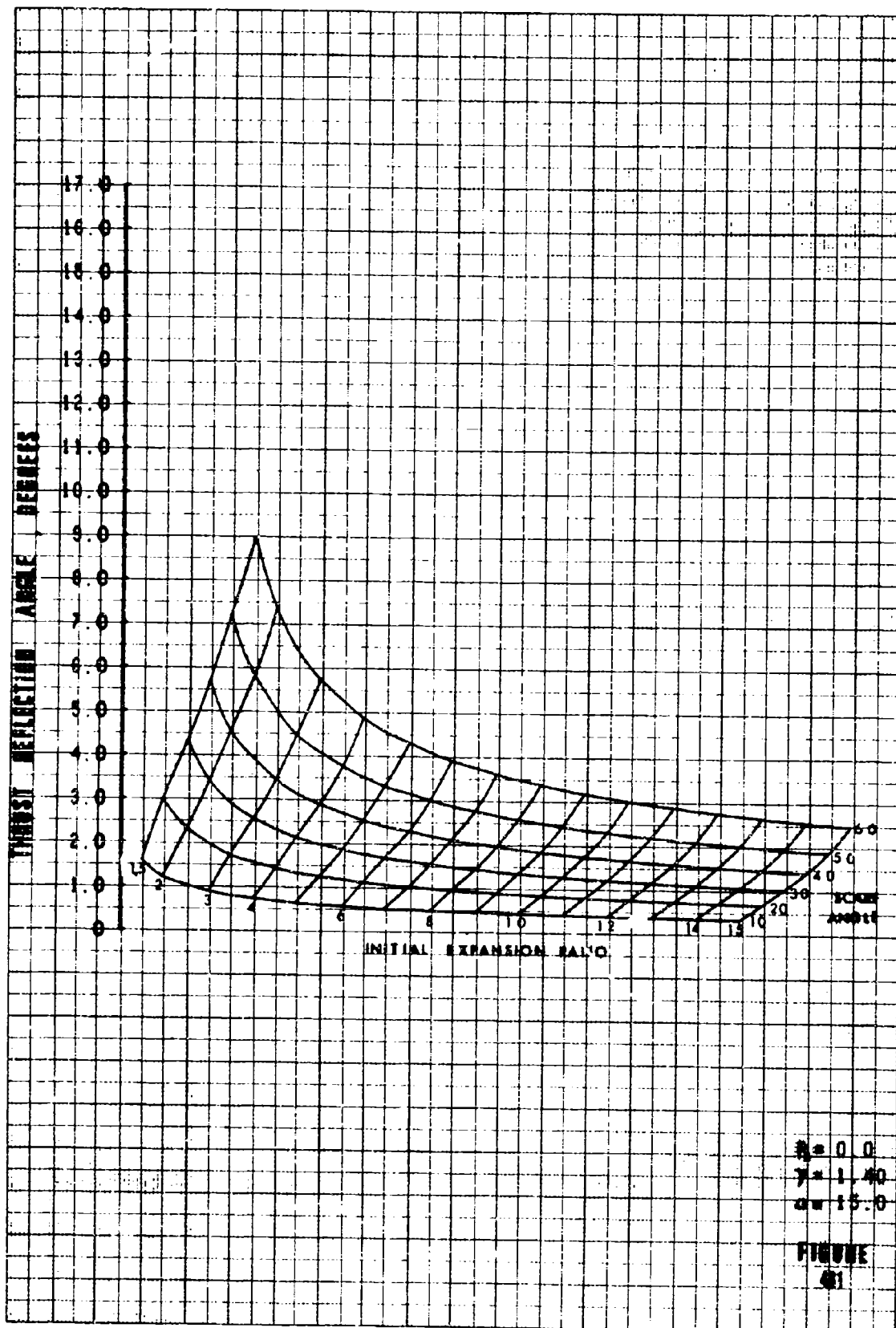












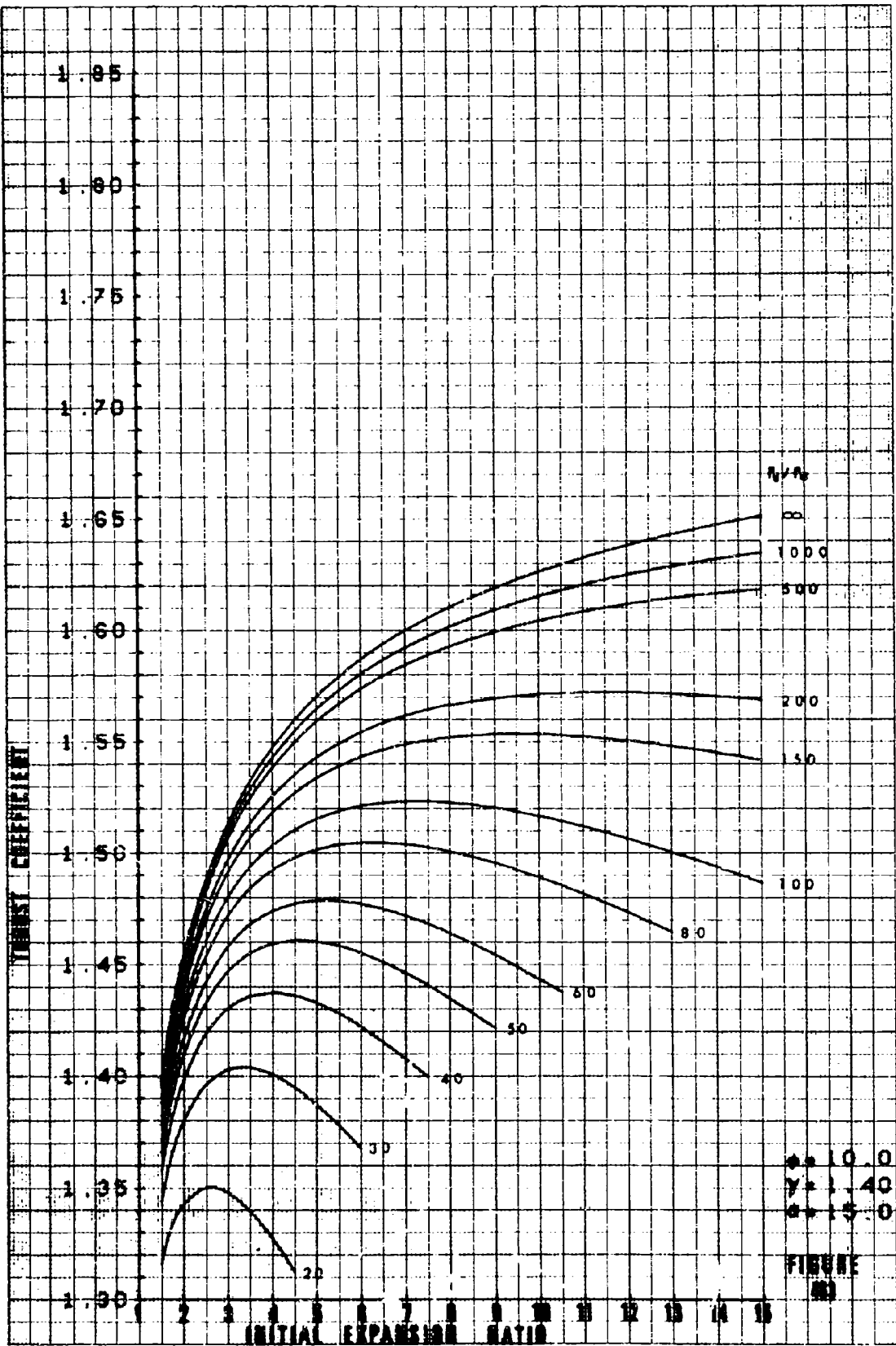


FIGURE 93

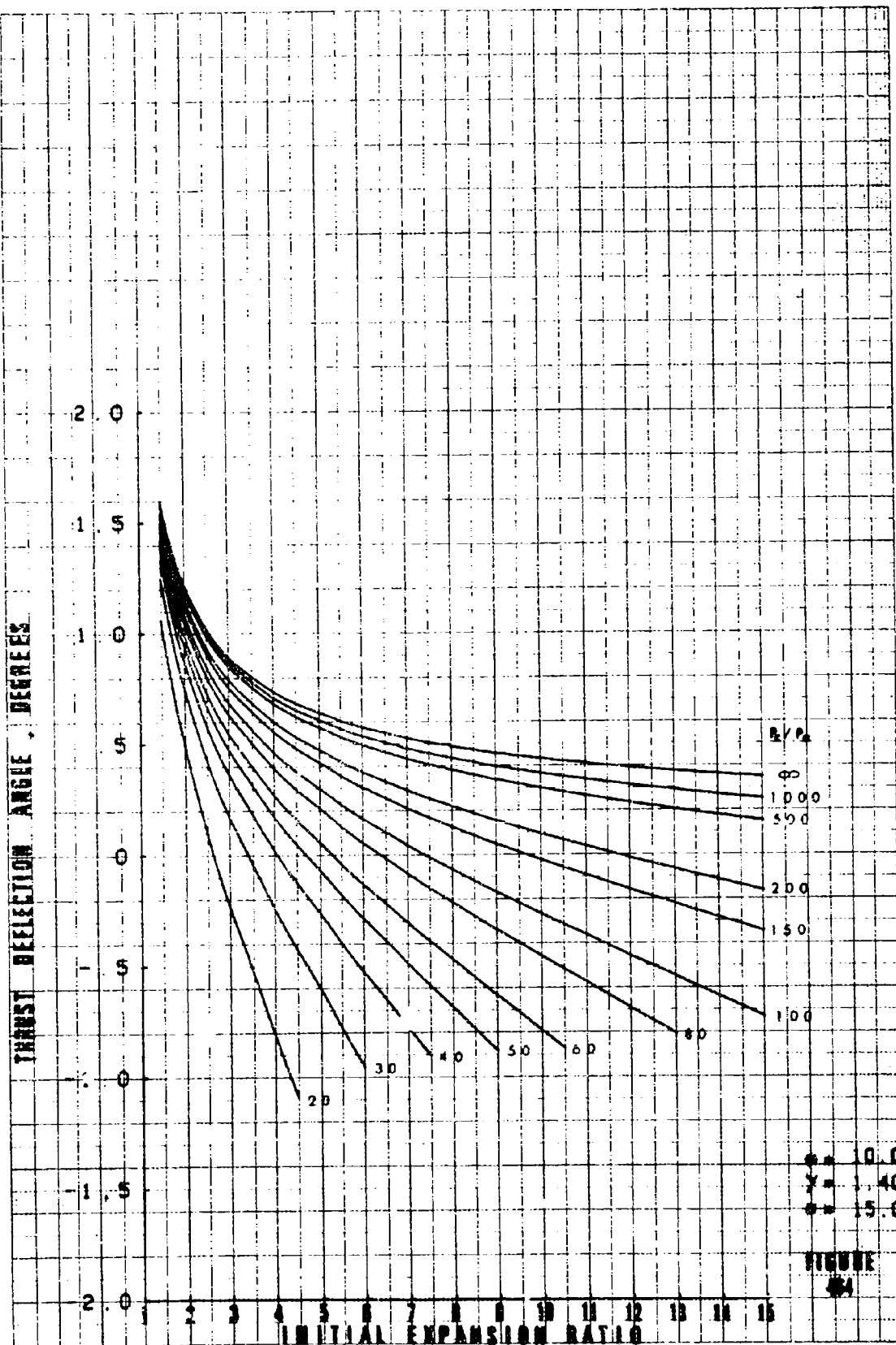
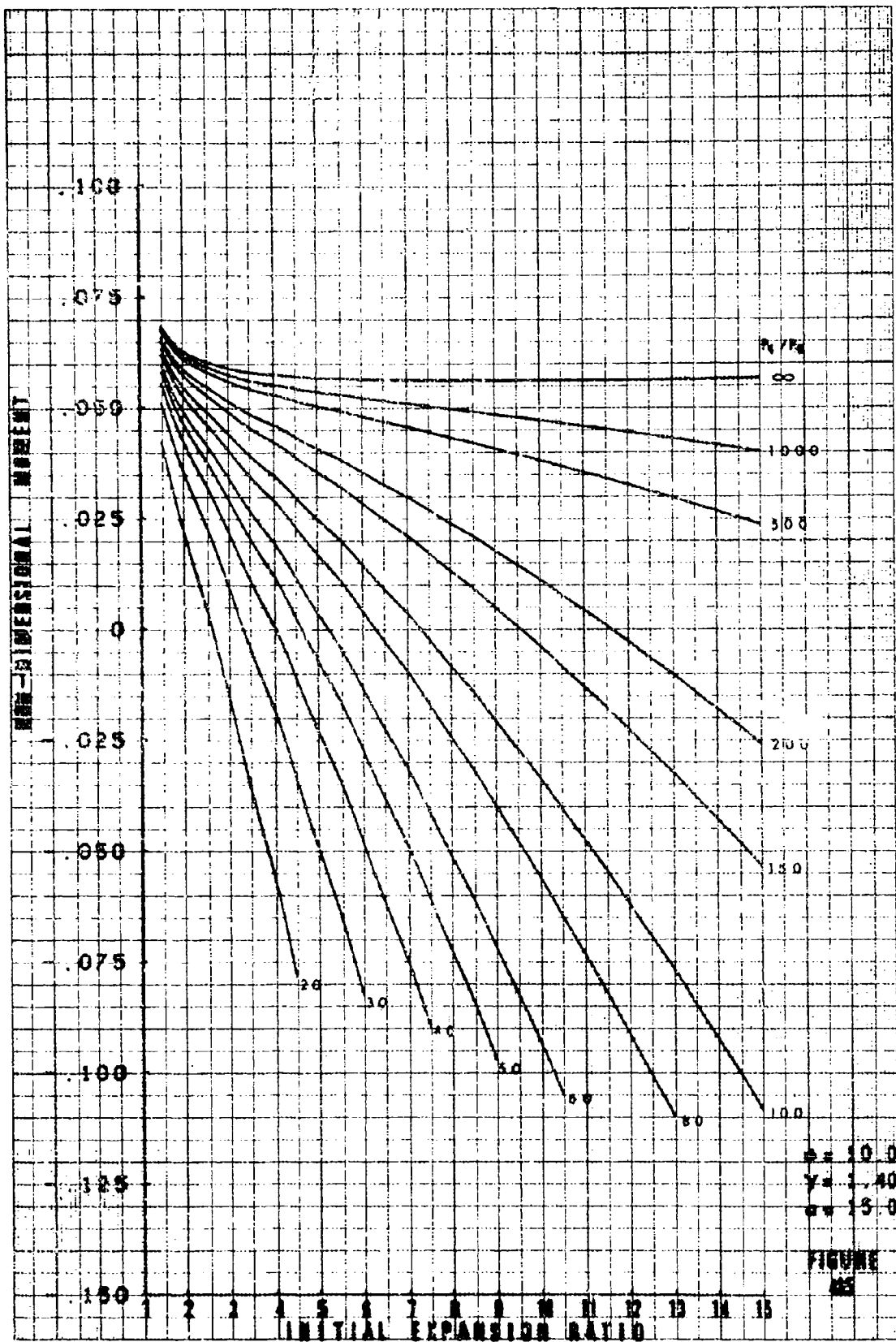
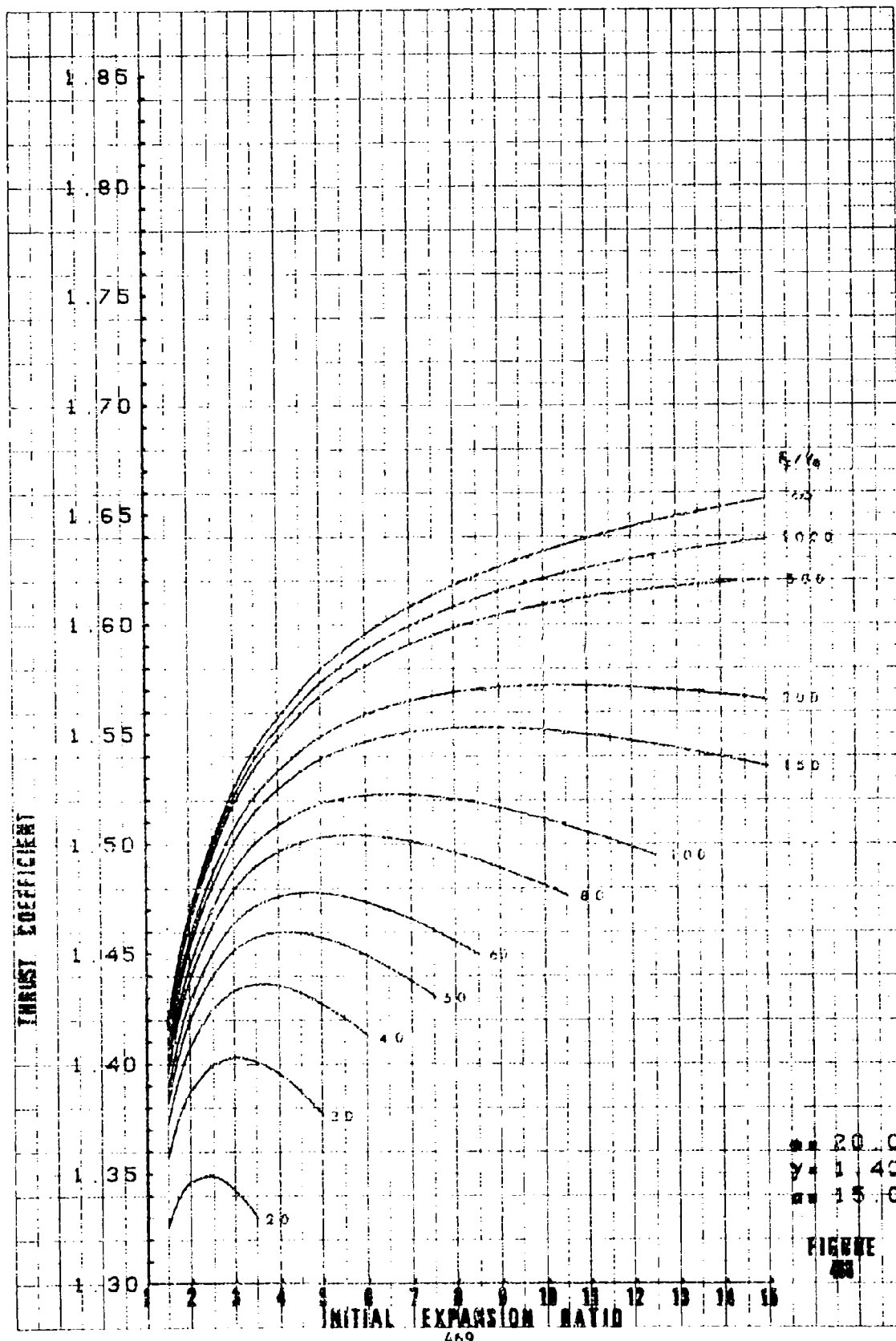
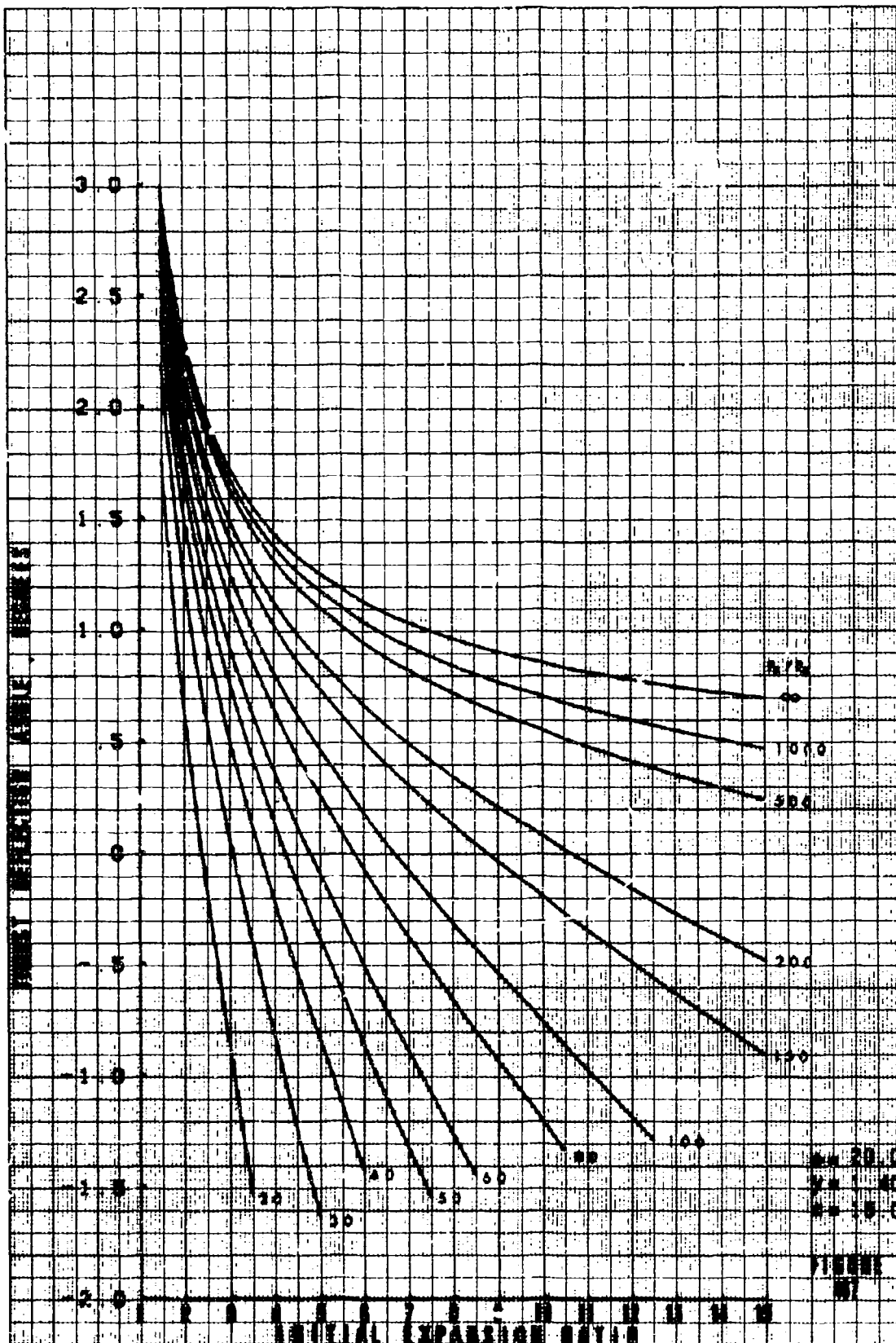
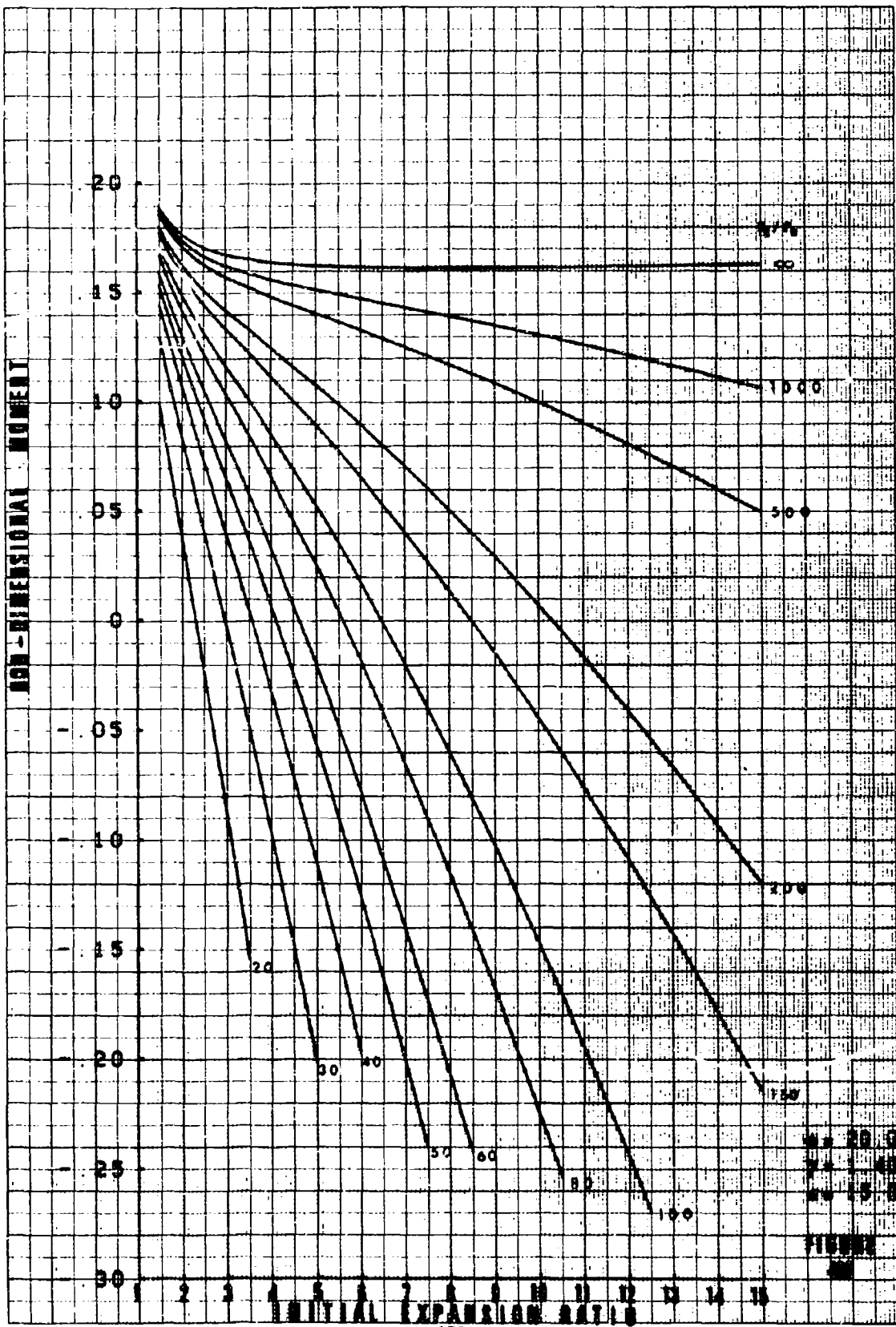


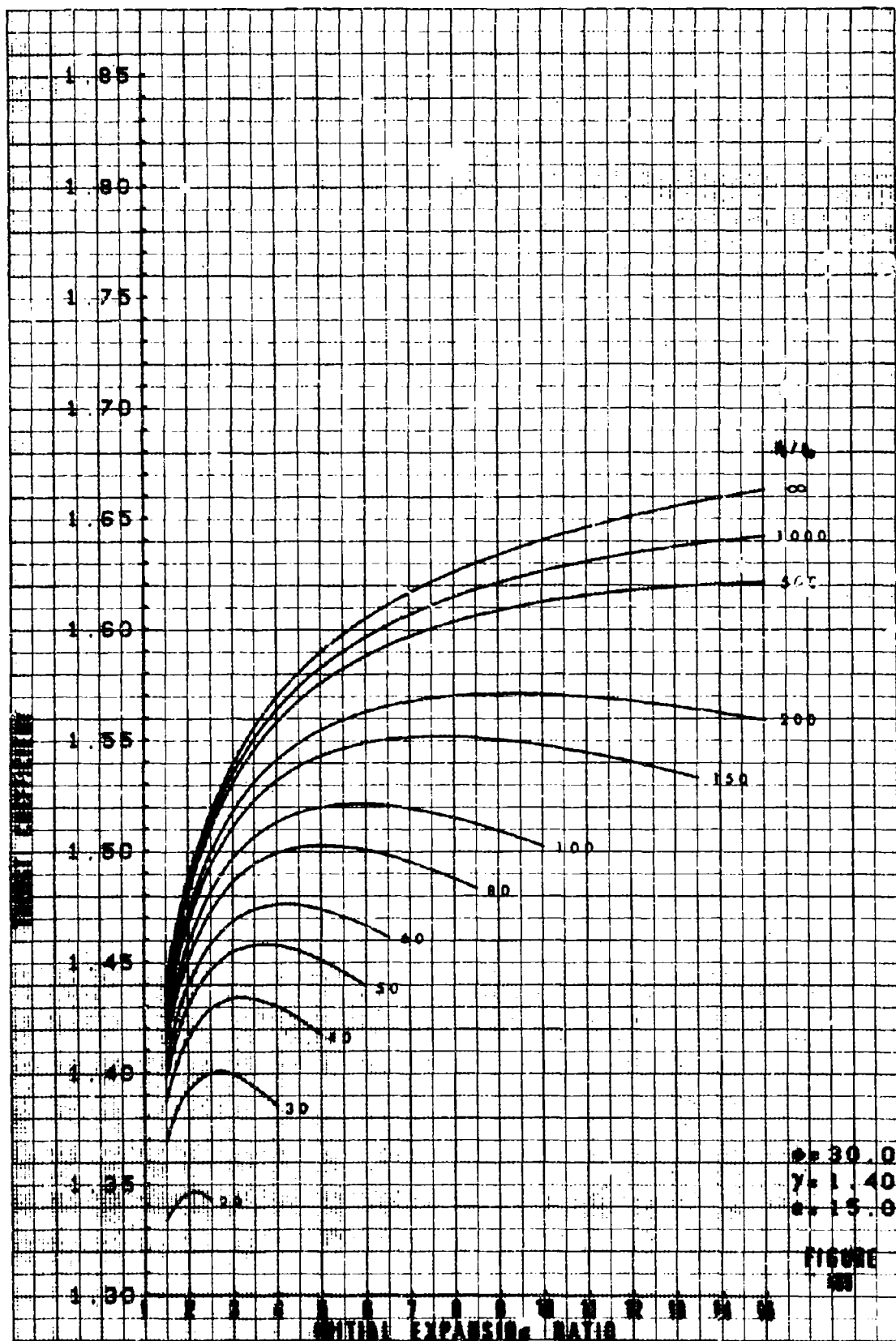
FIGURE 4

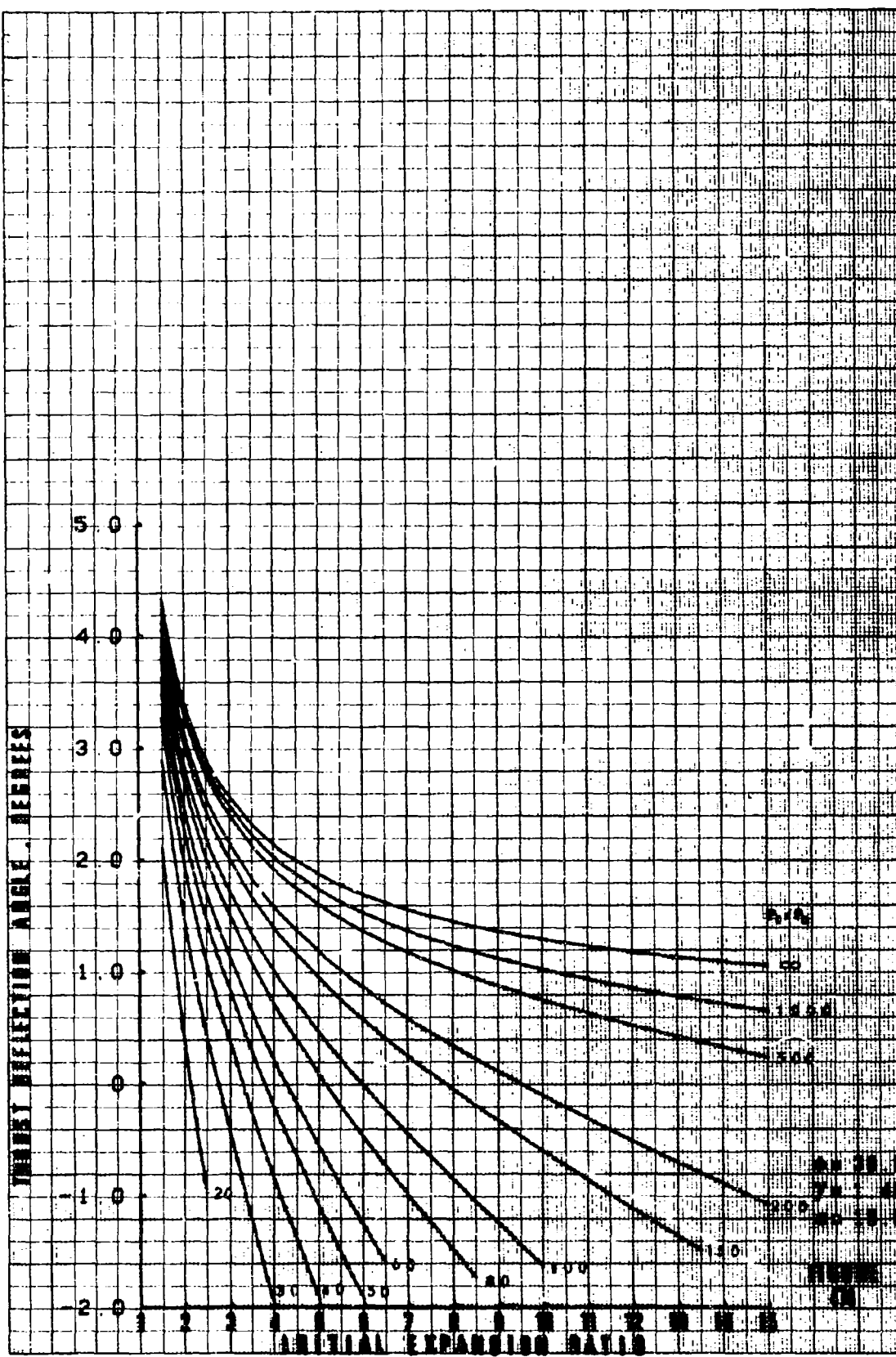


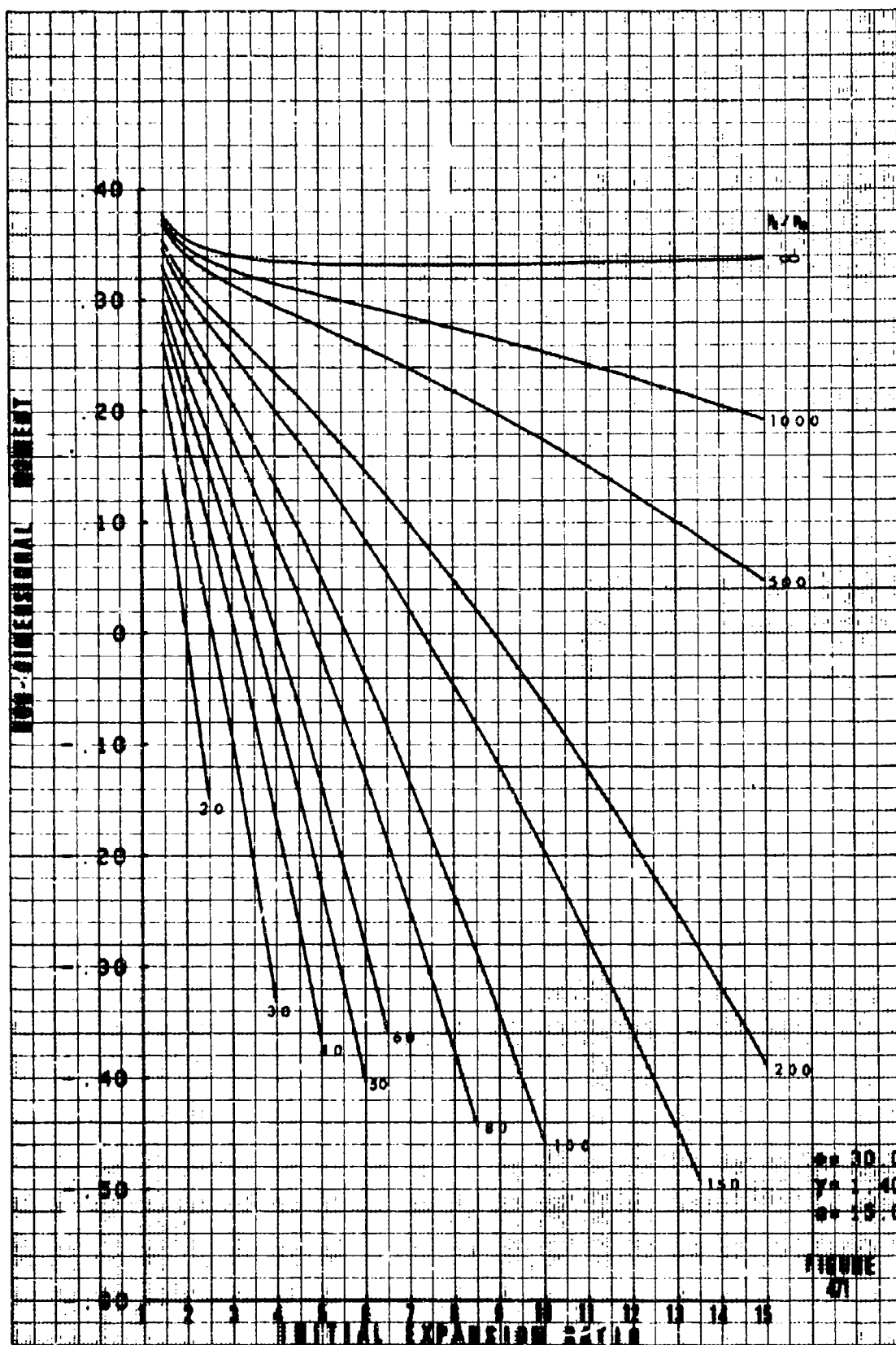


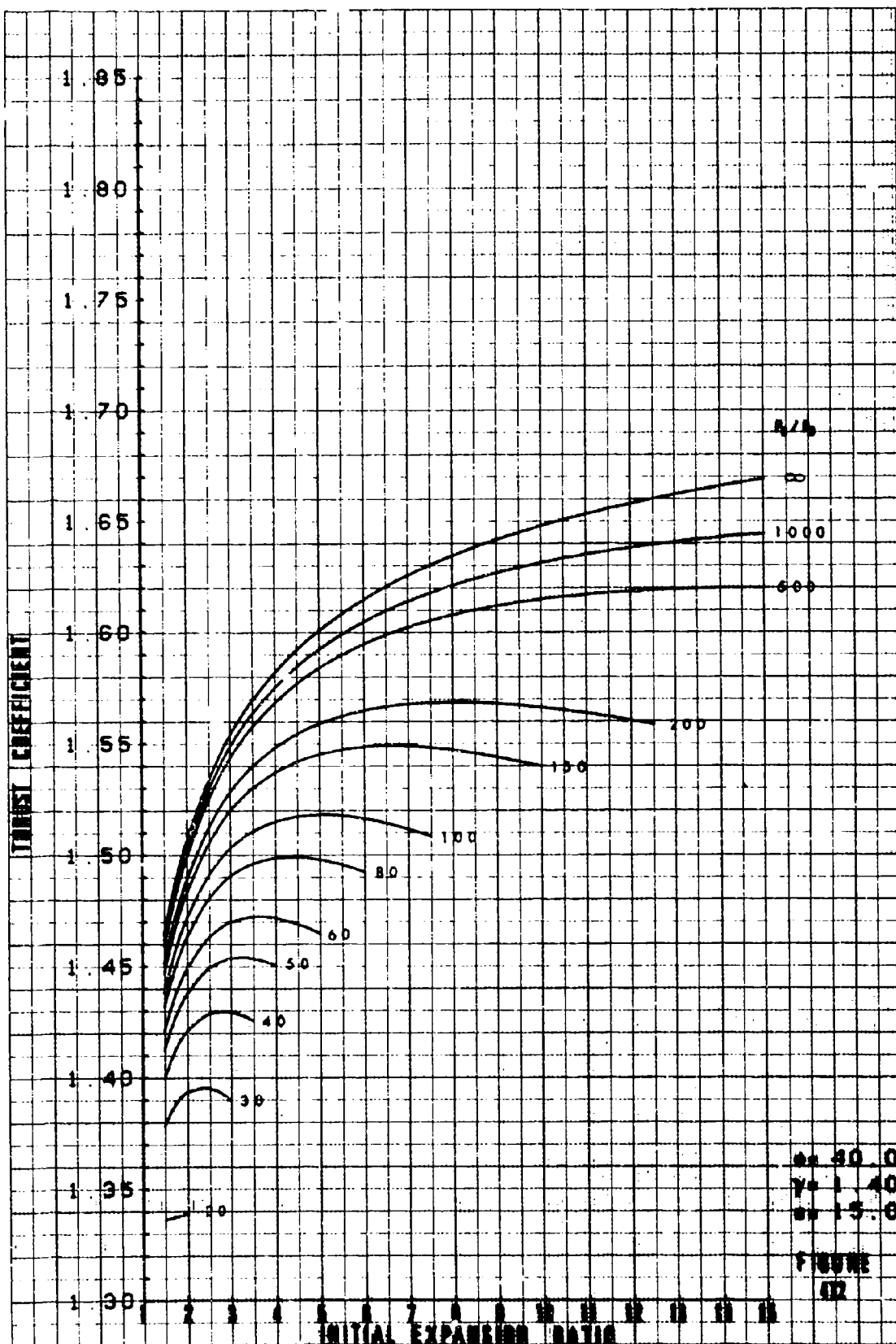






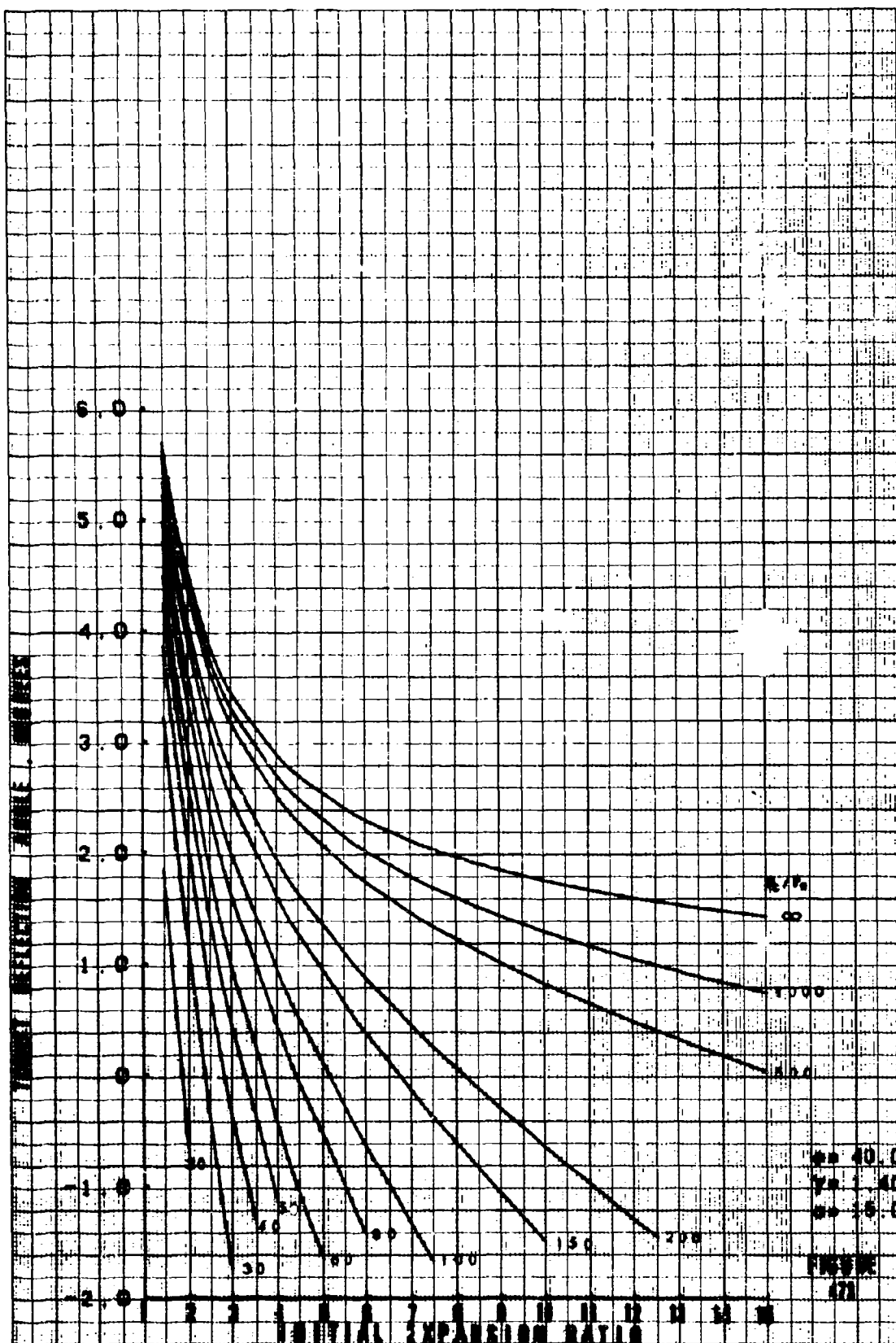


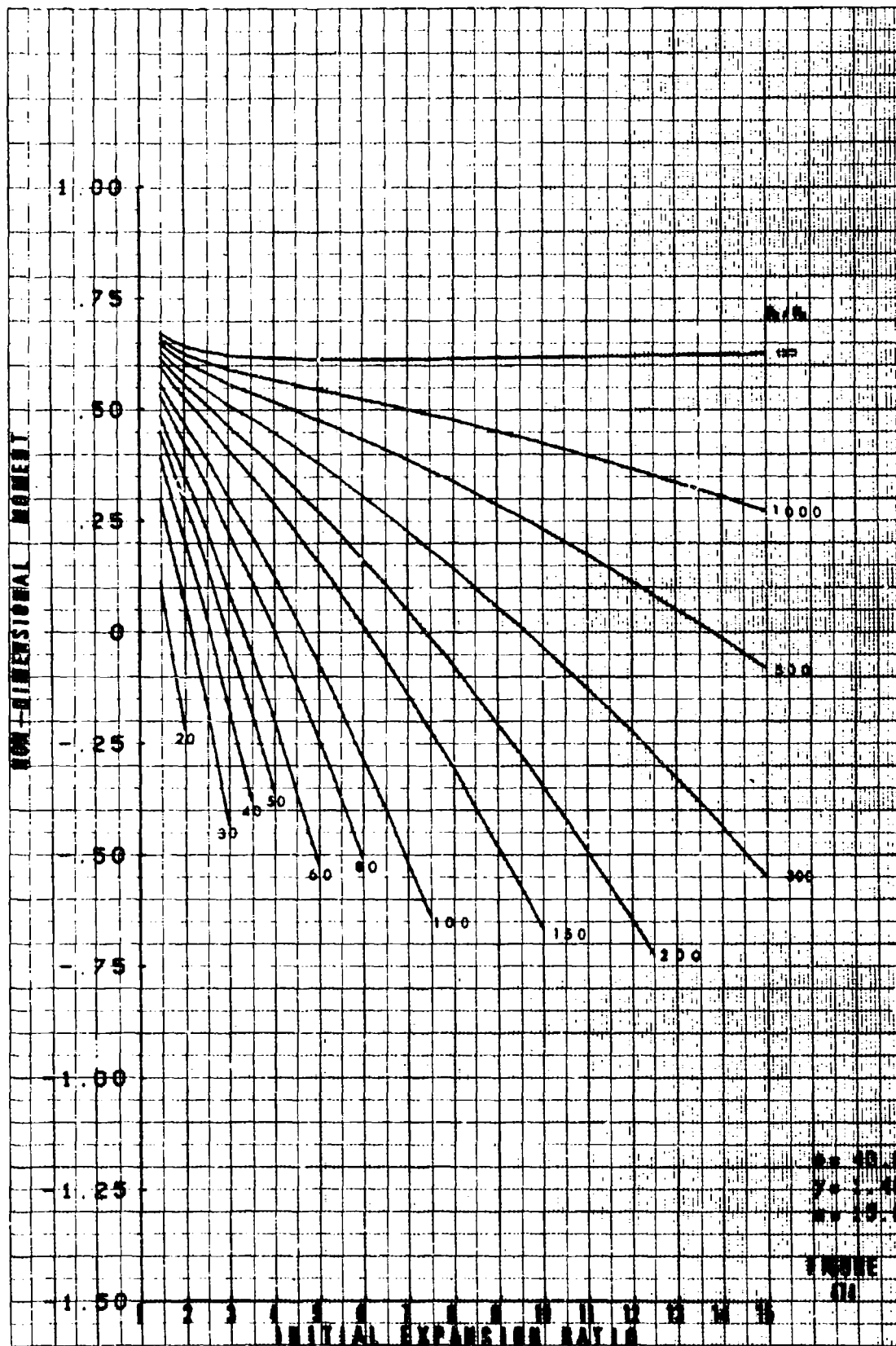


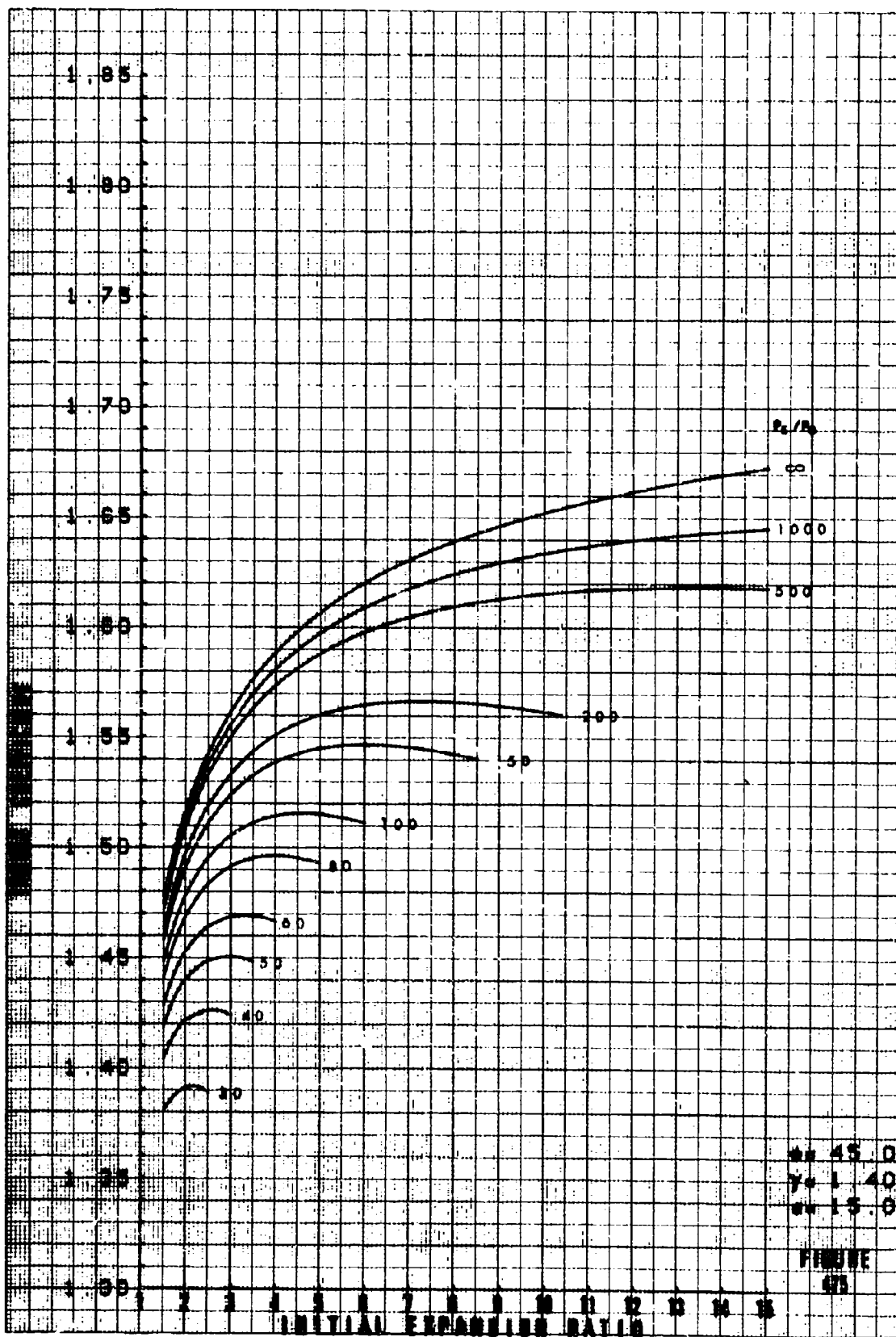


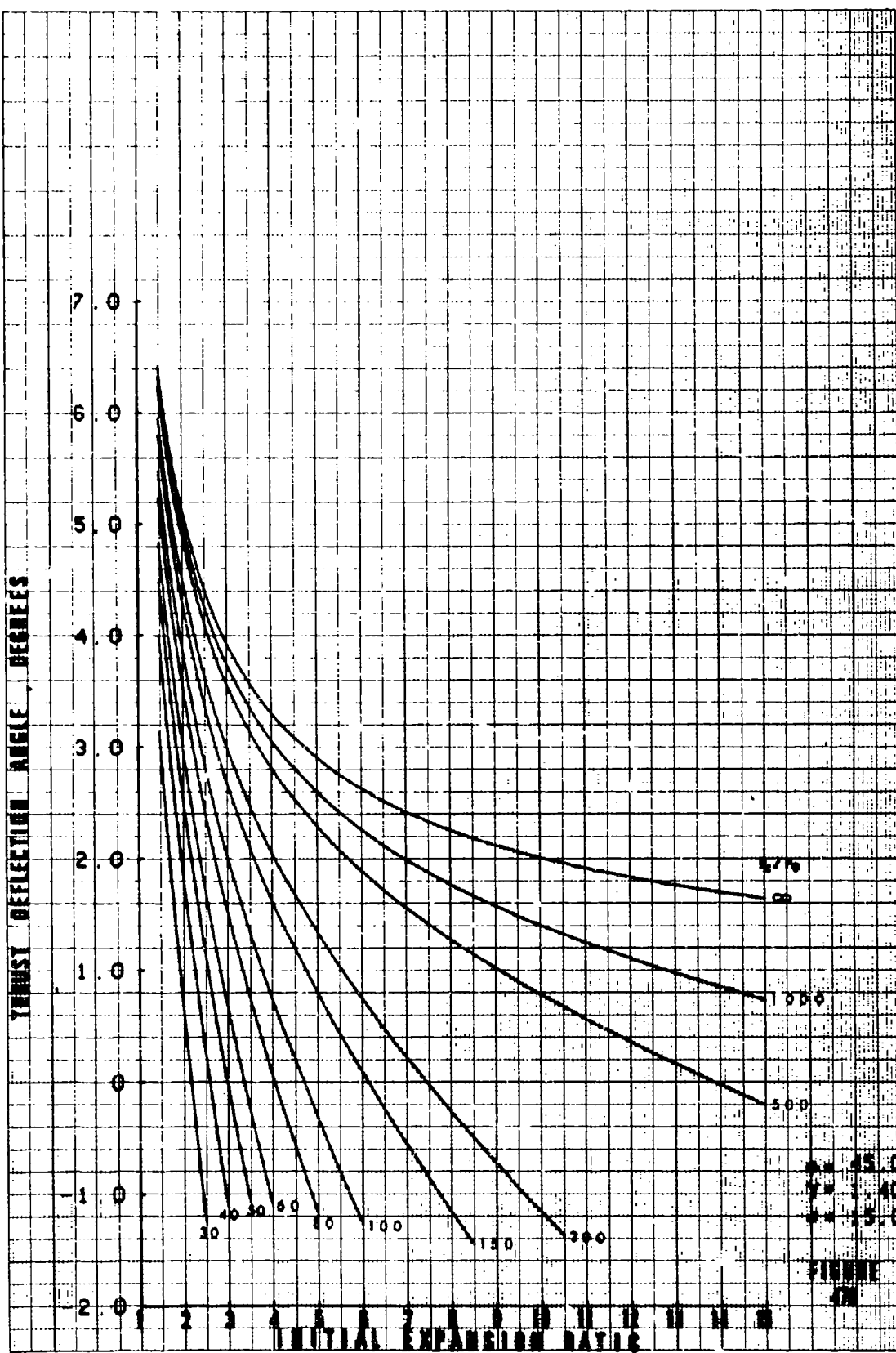
$\gamma = 1.40$
 $\gamma = 1.50$

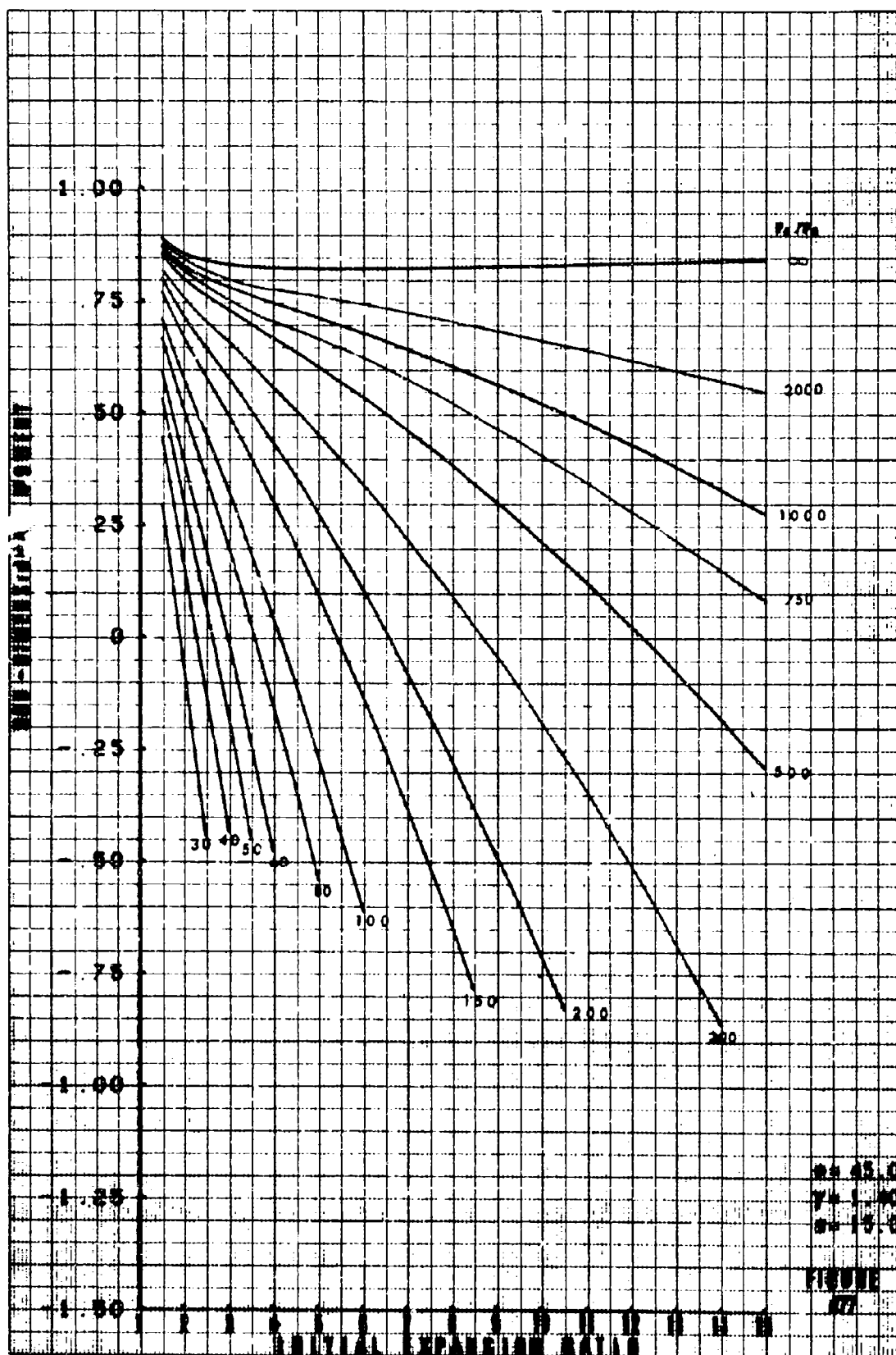
FIGURE 12

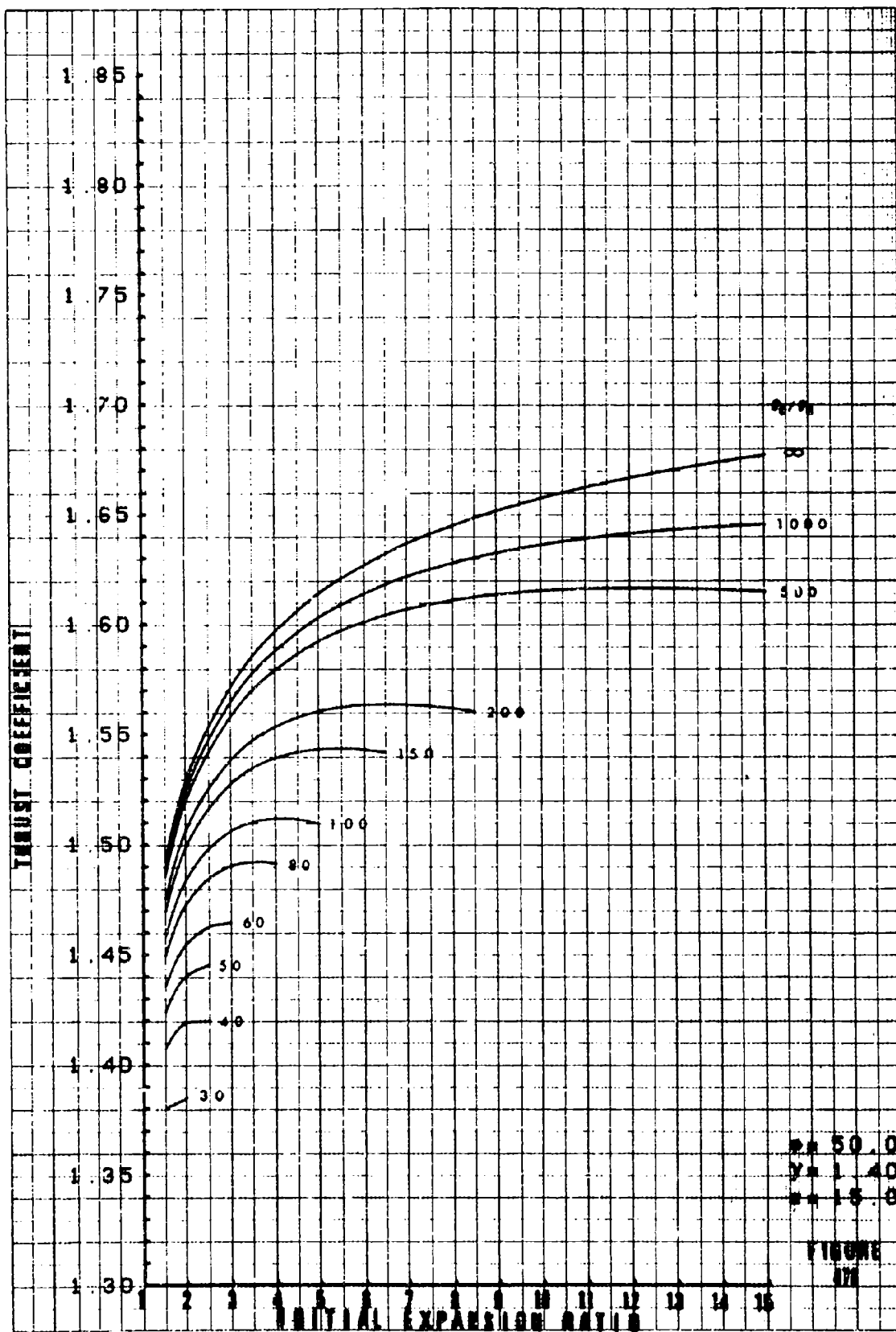












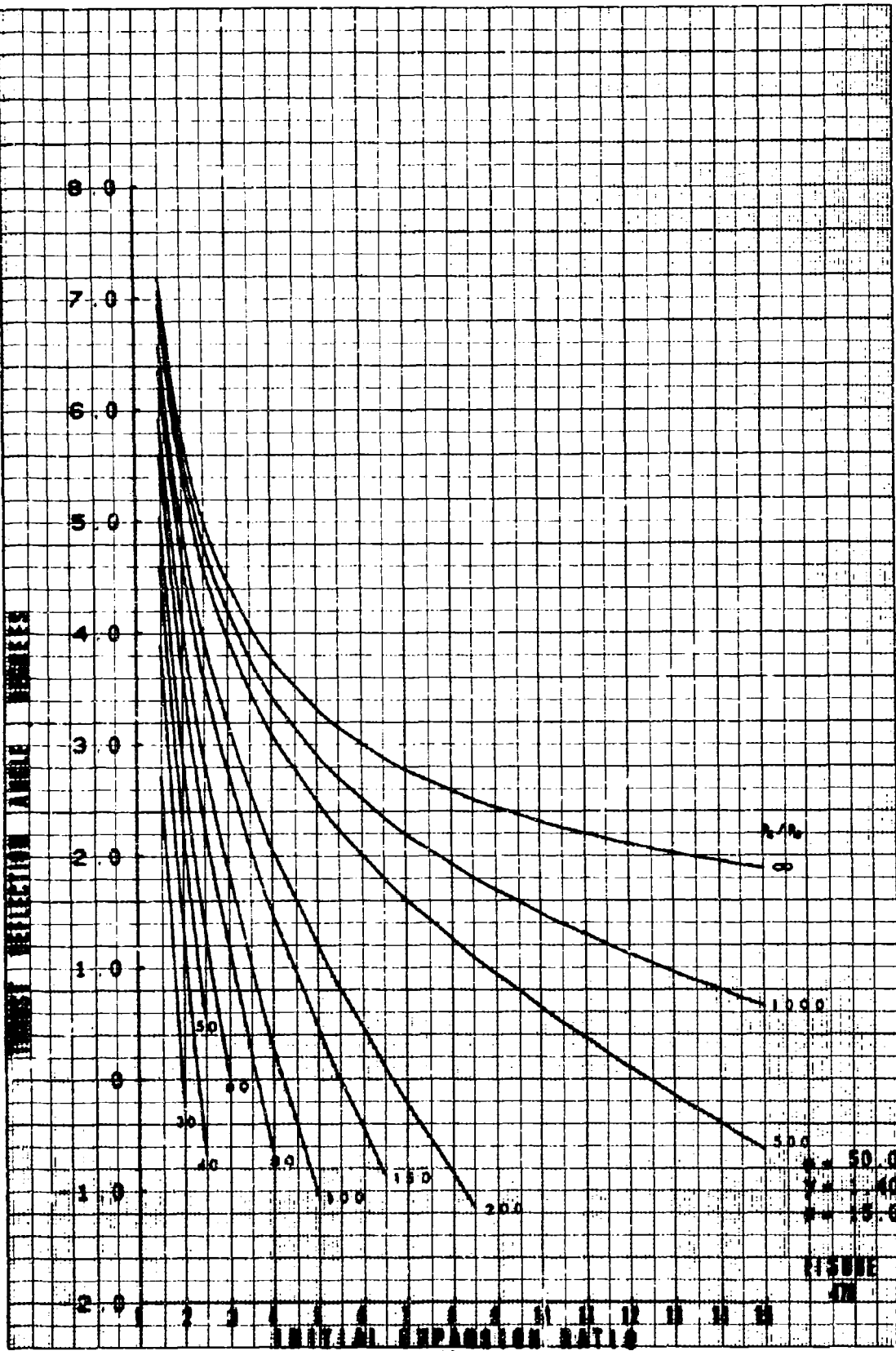
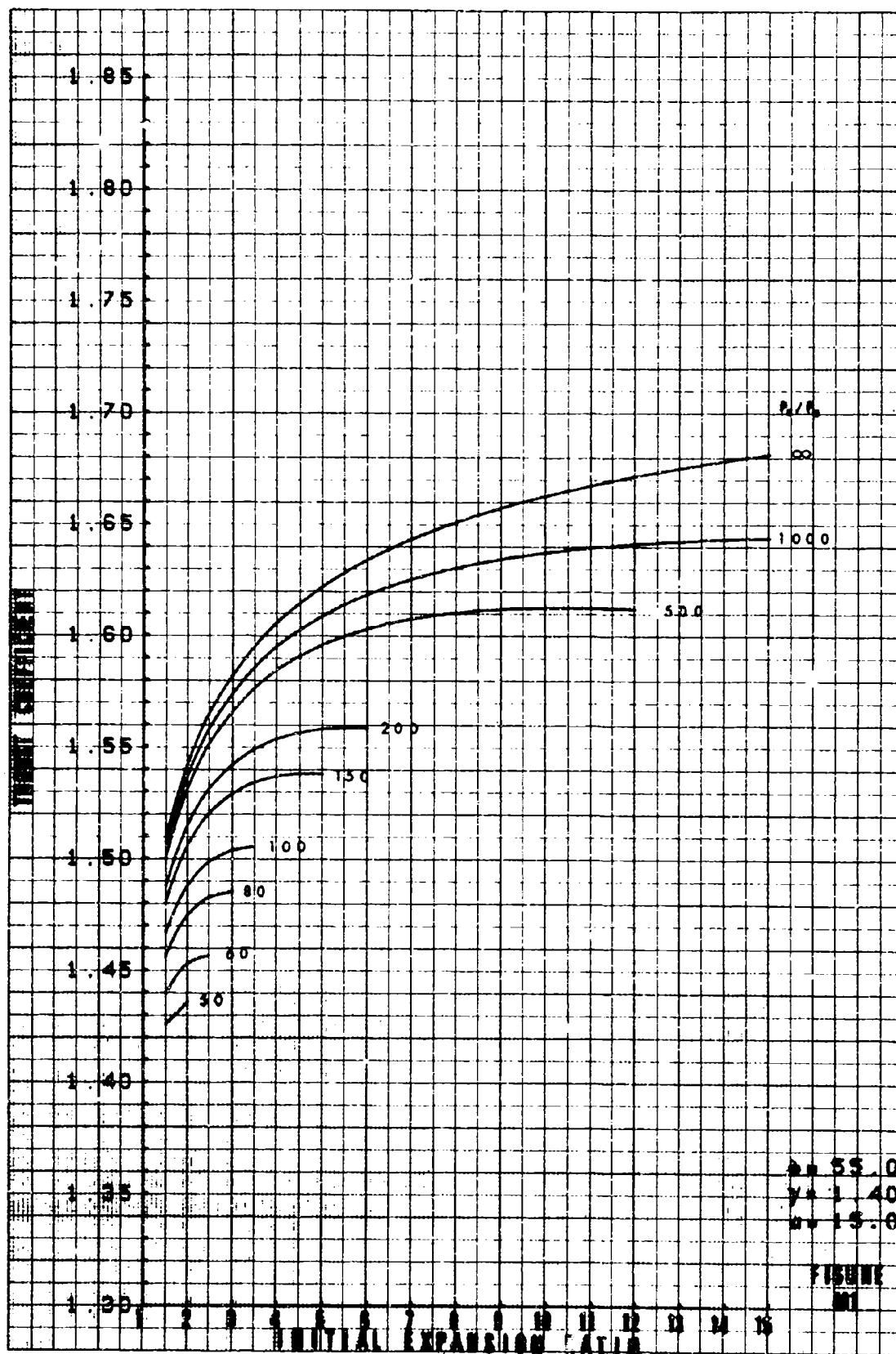




FIGURE 2



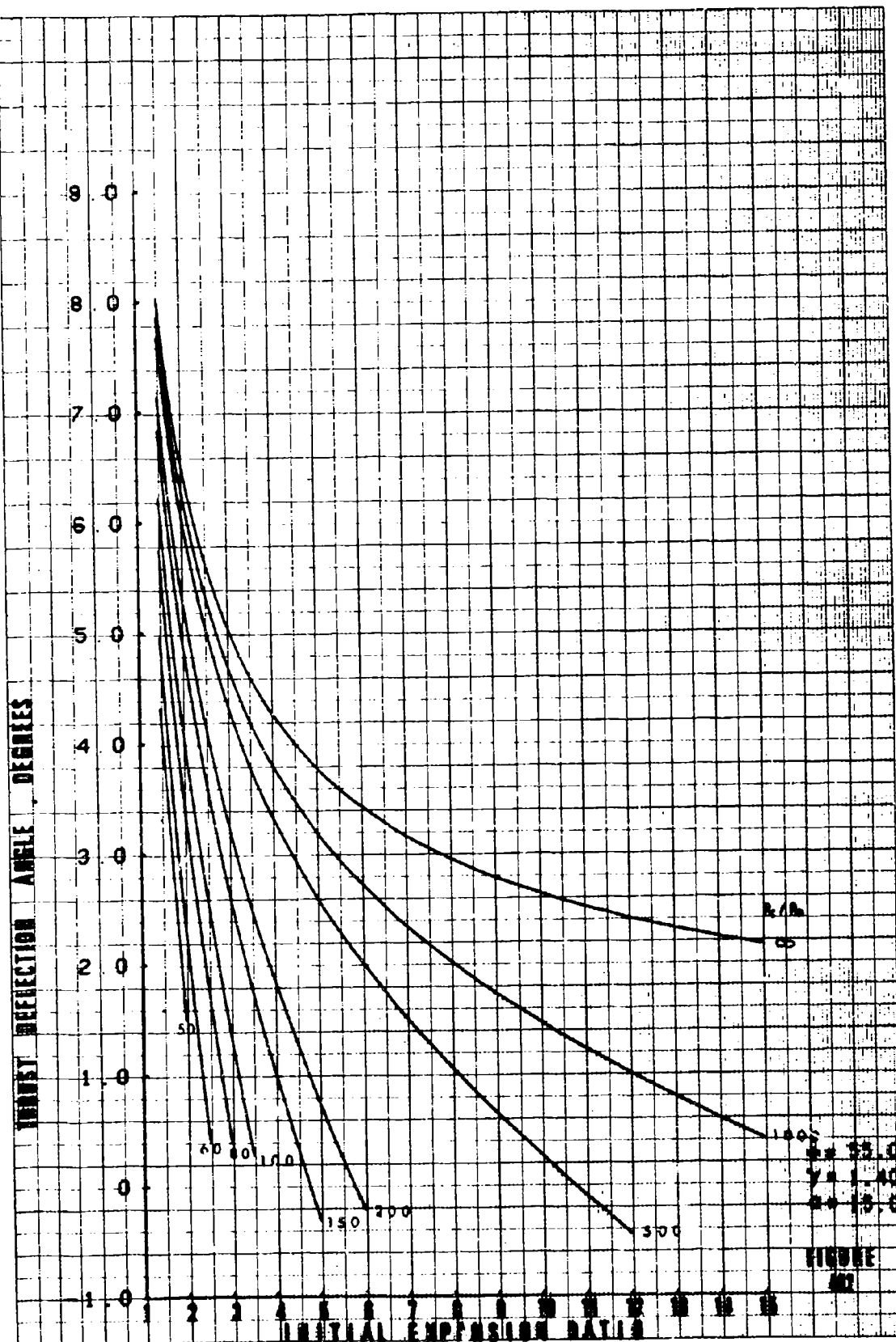
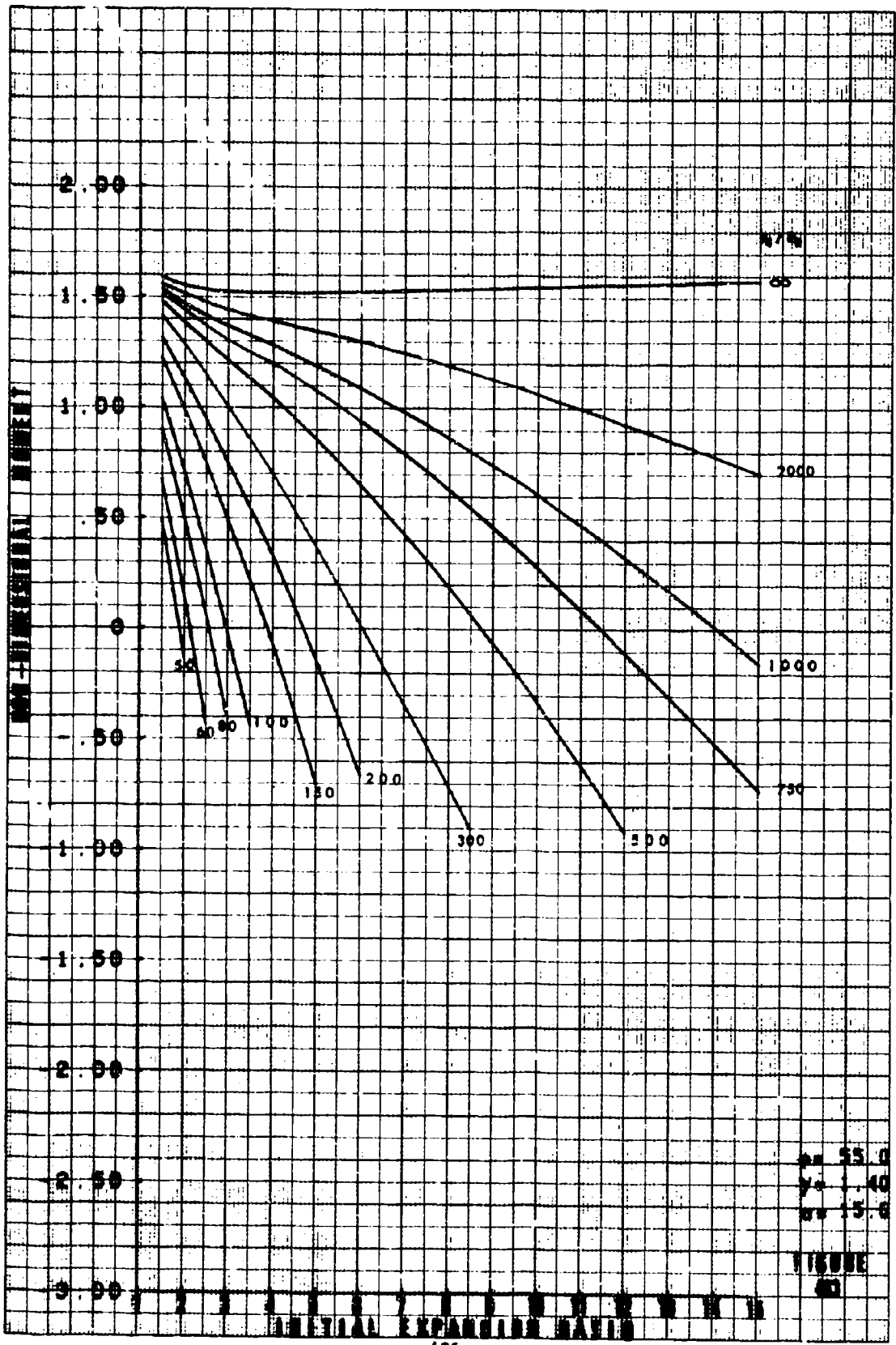


FIGURE 1



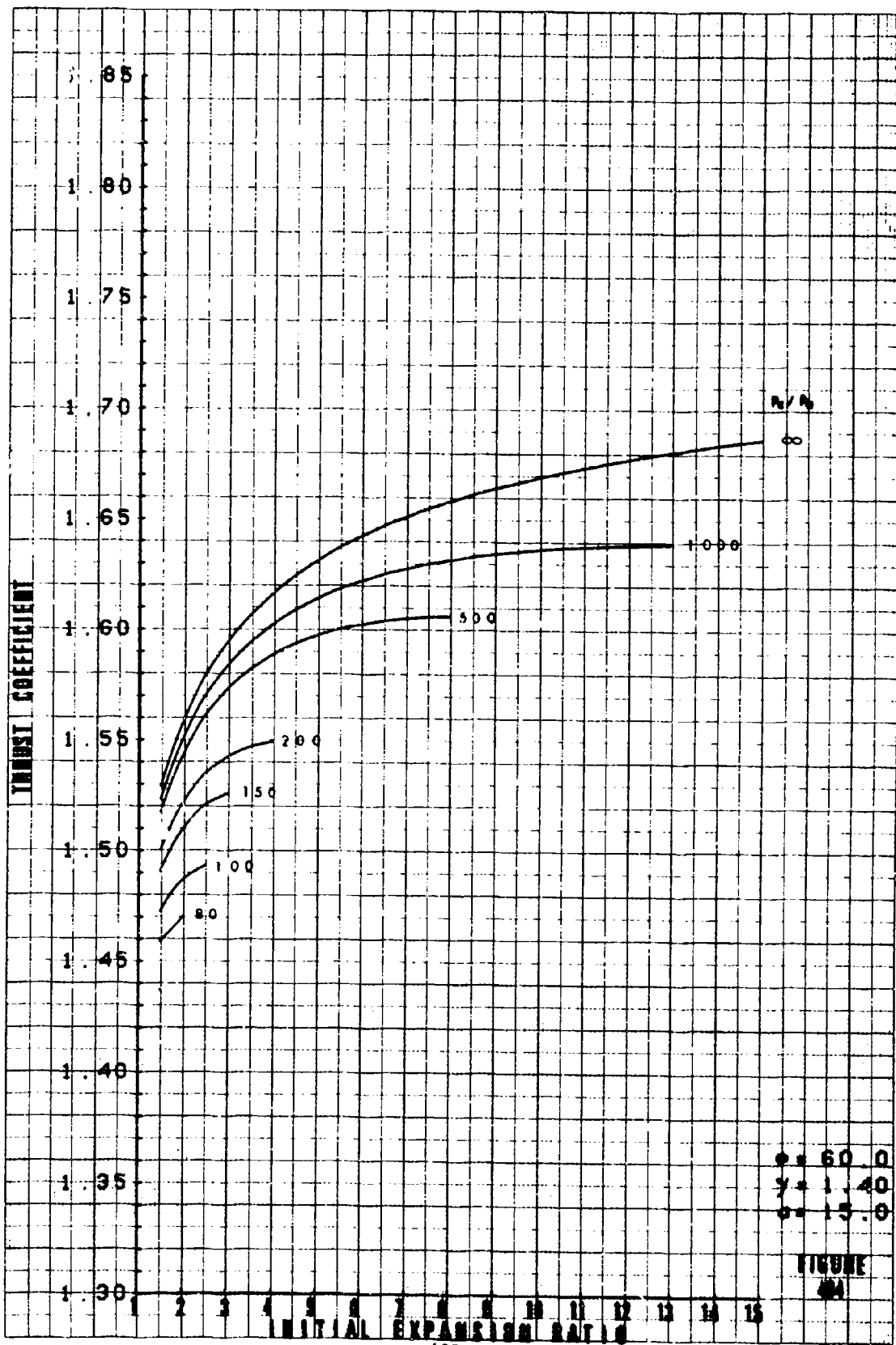
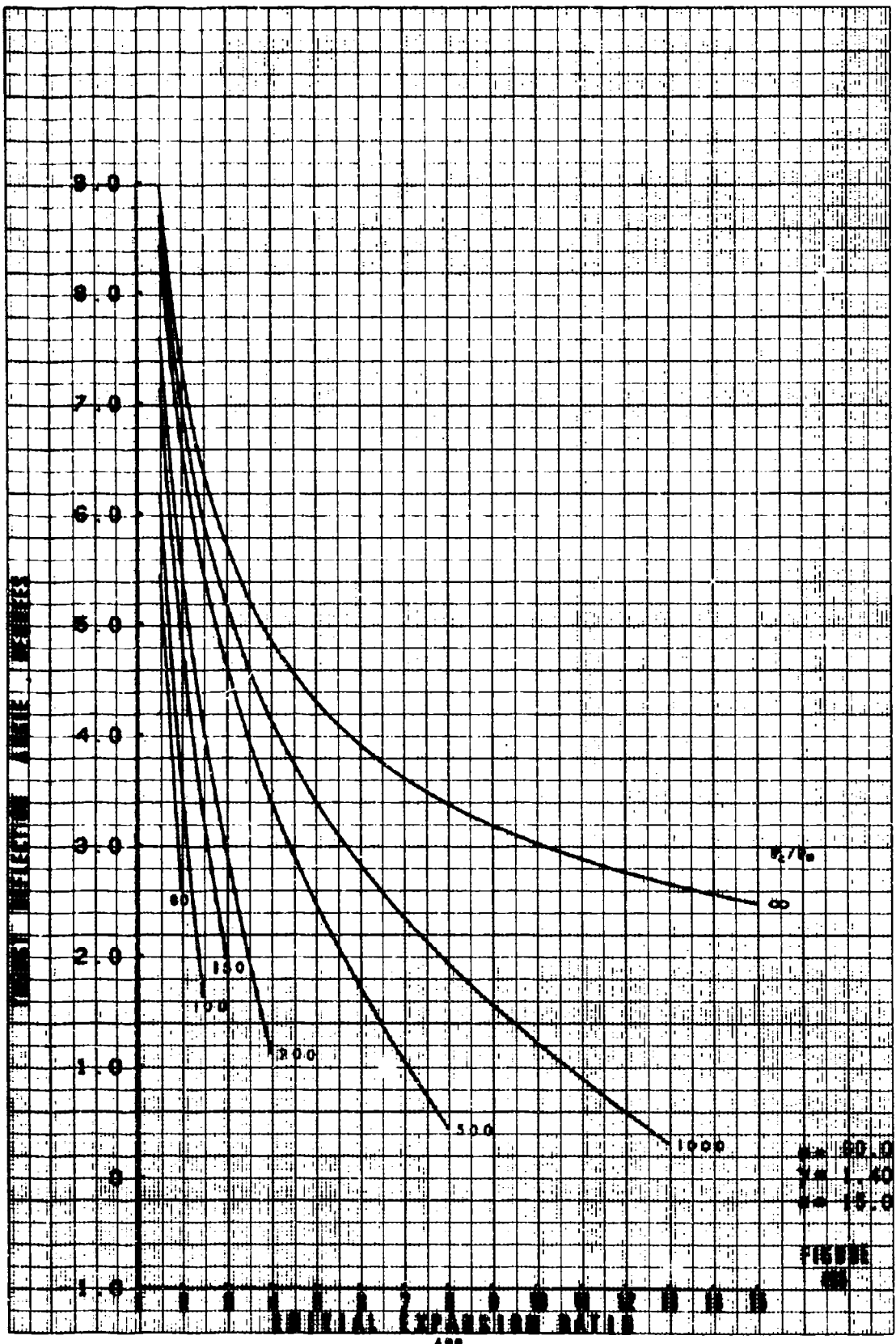
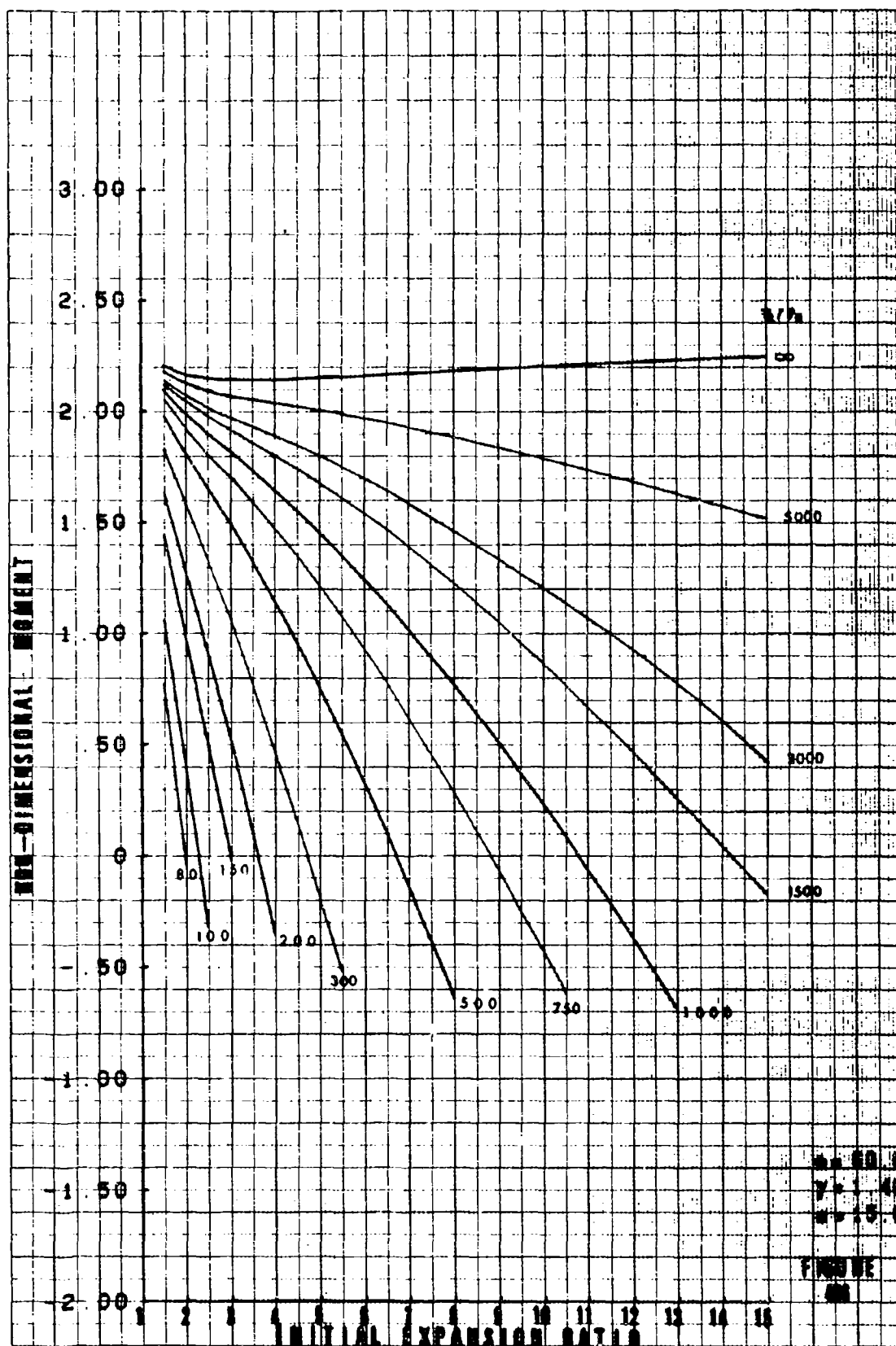
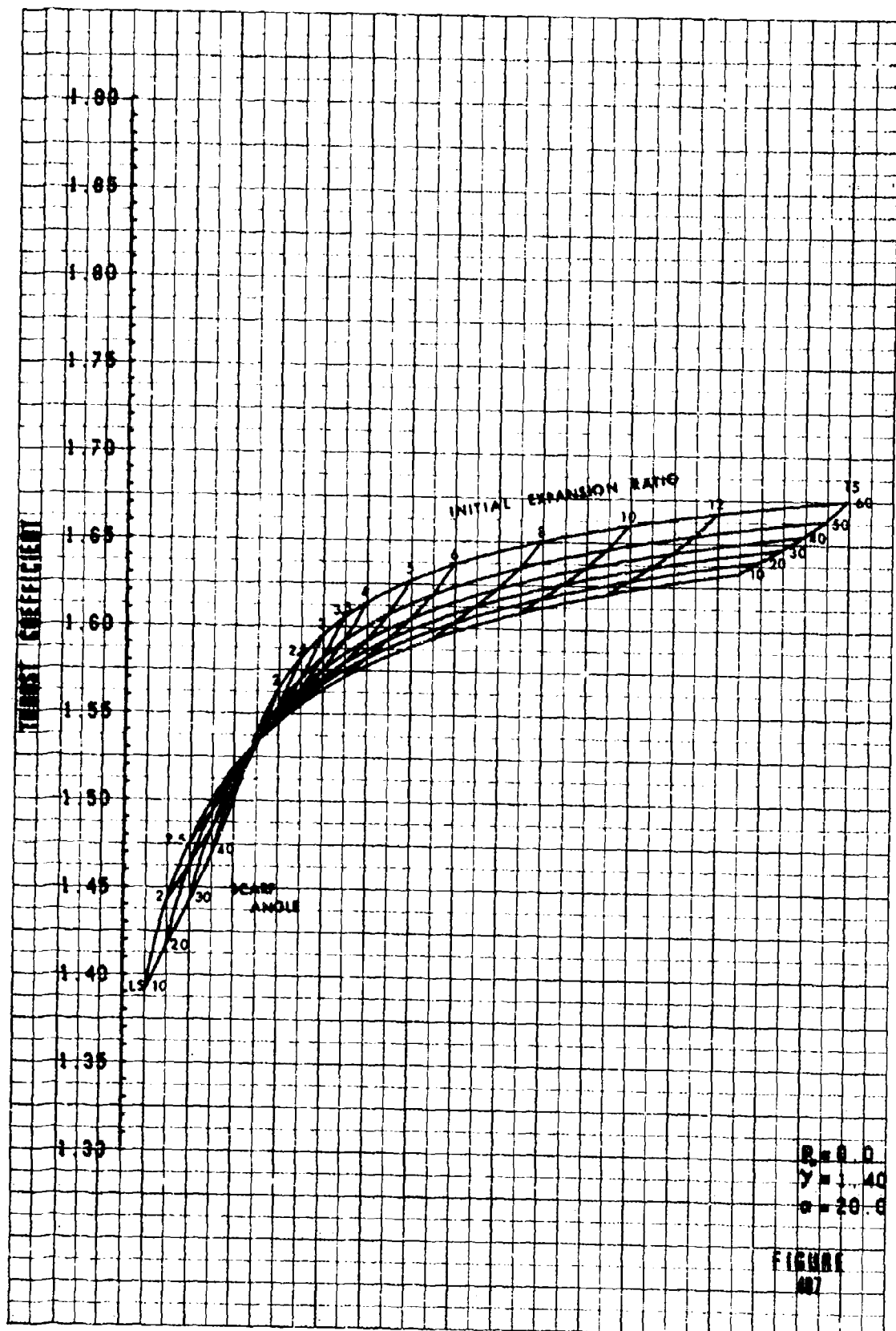
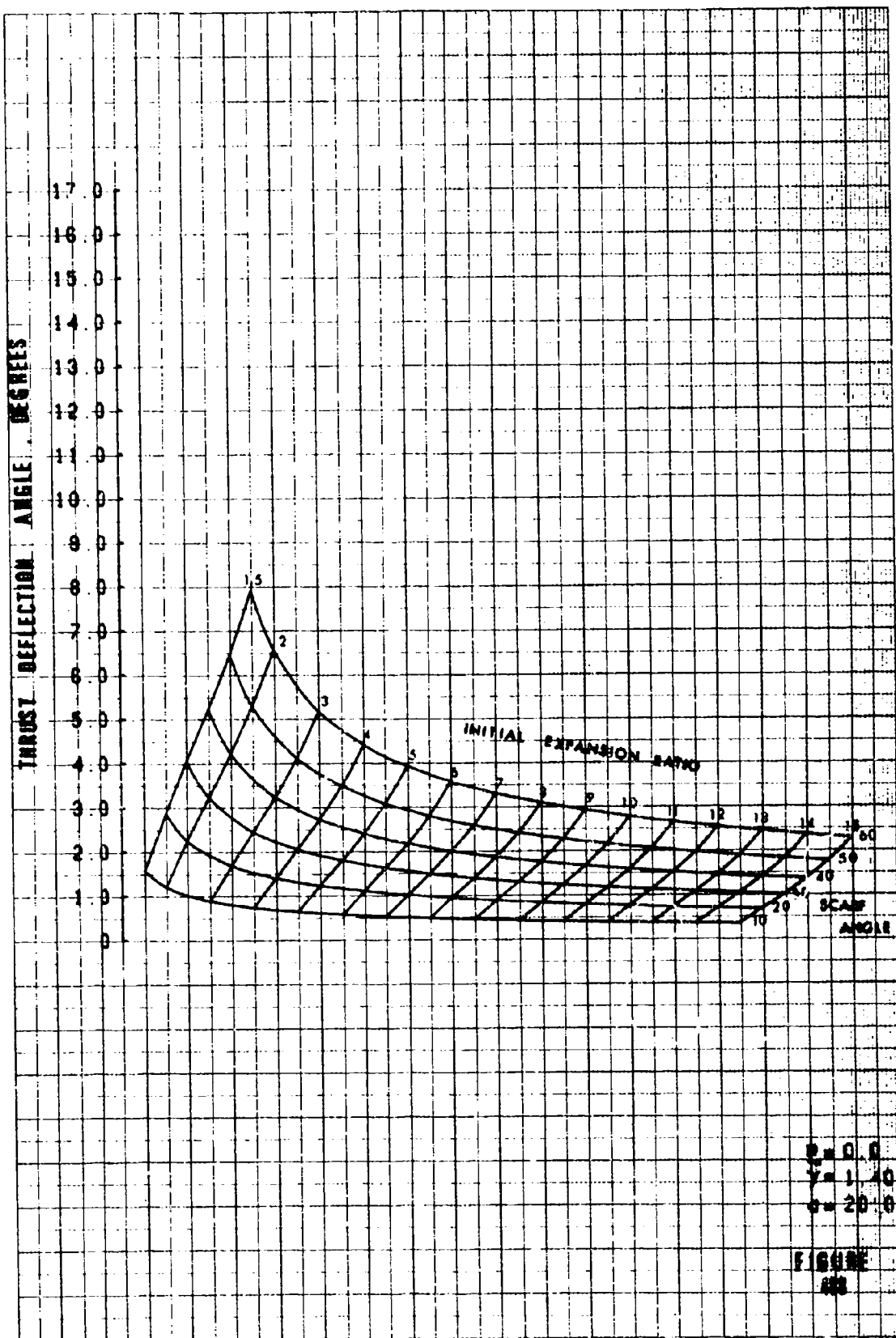


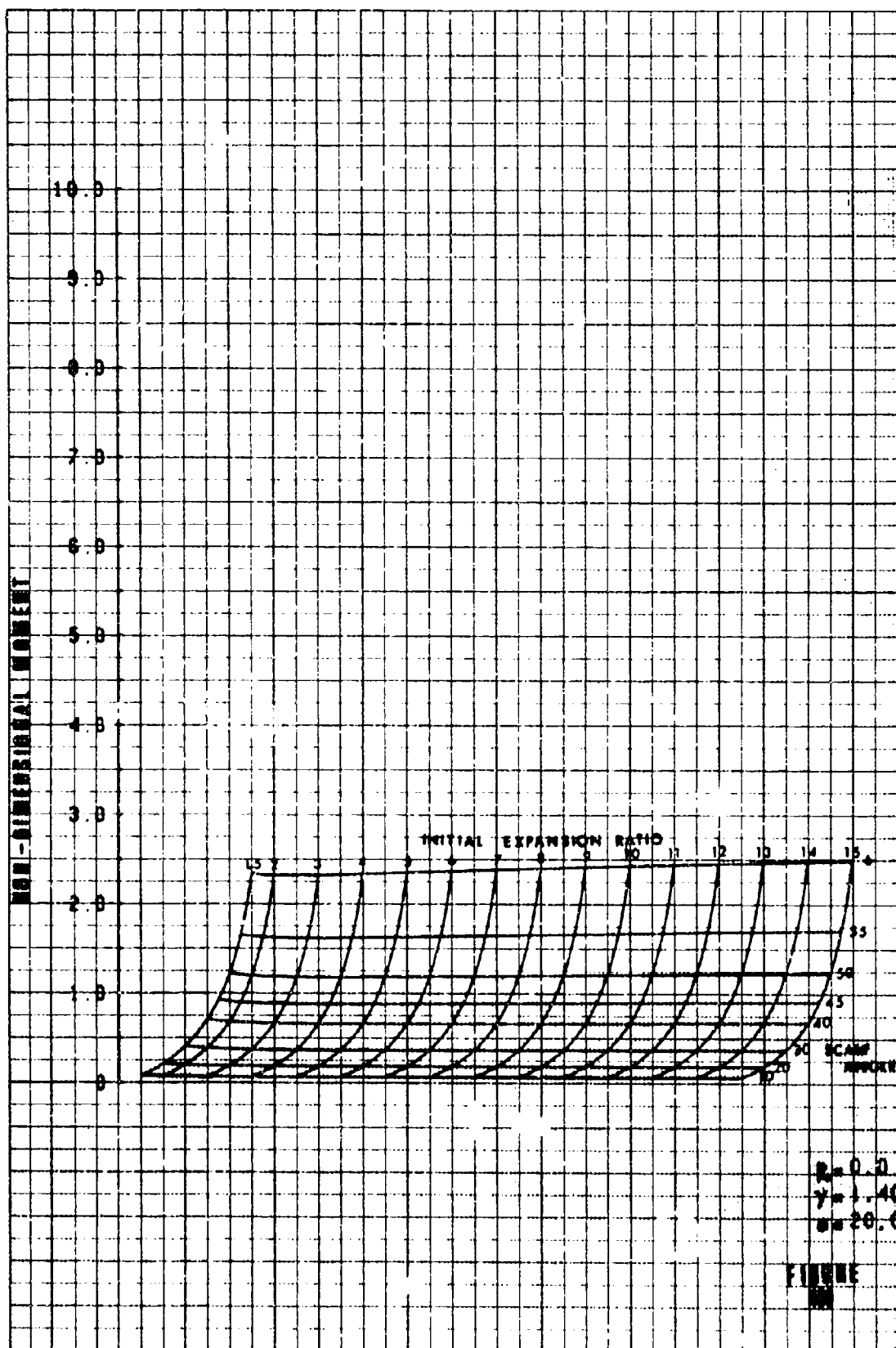
FIGURE 44

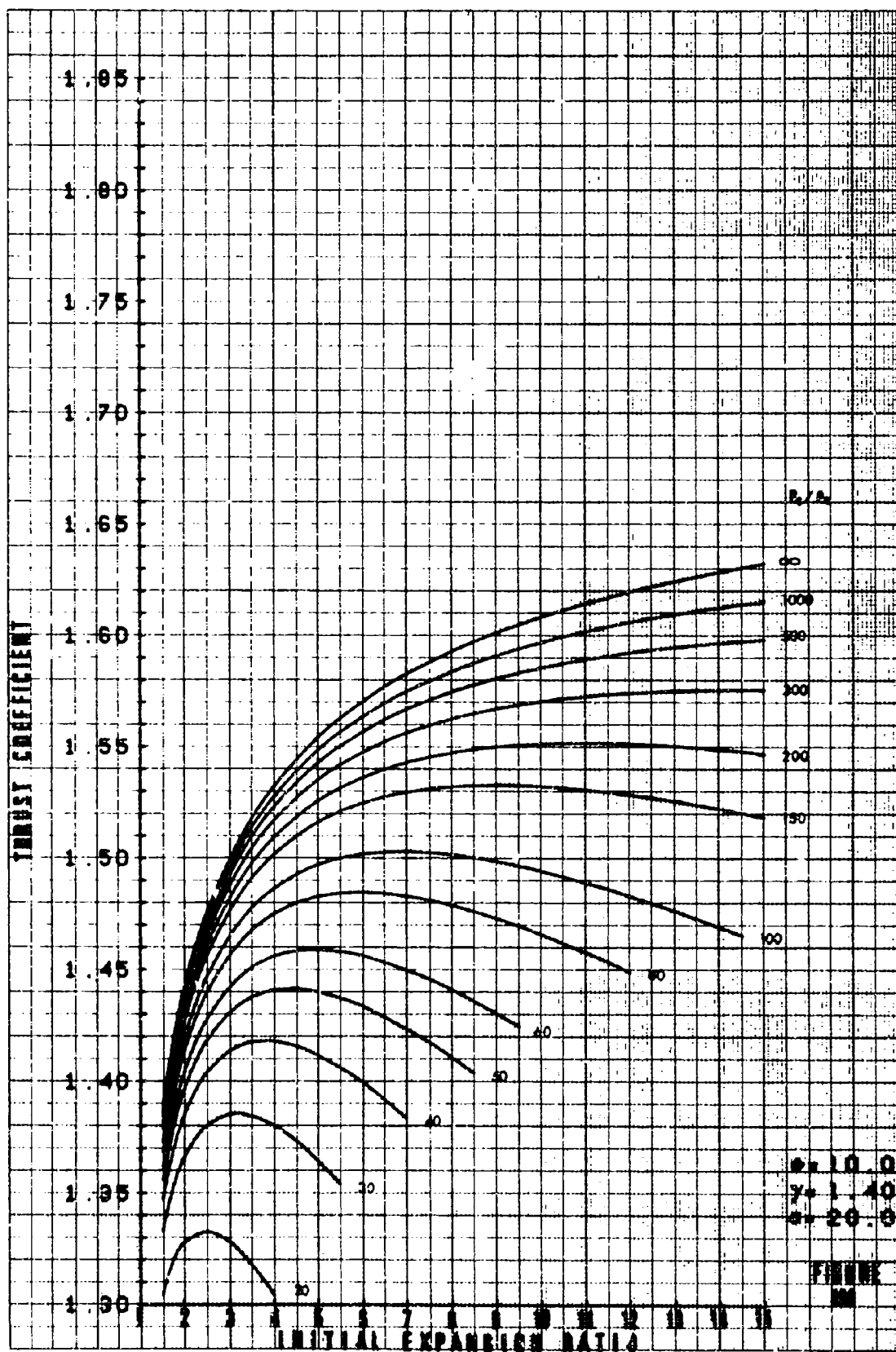


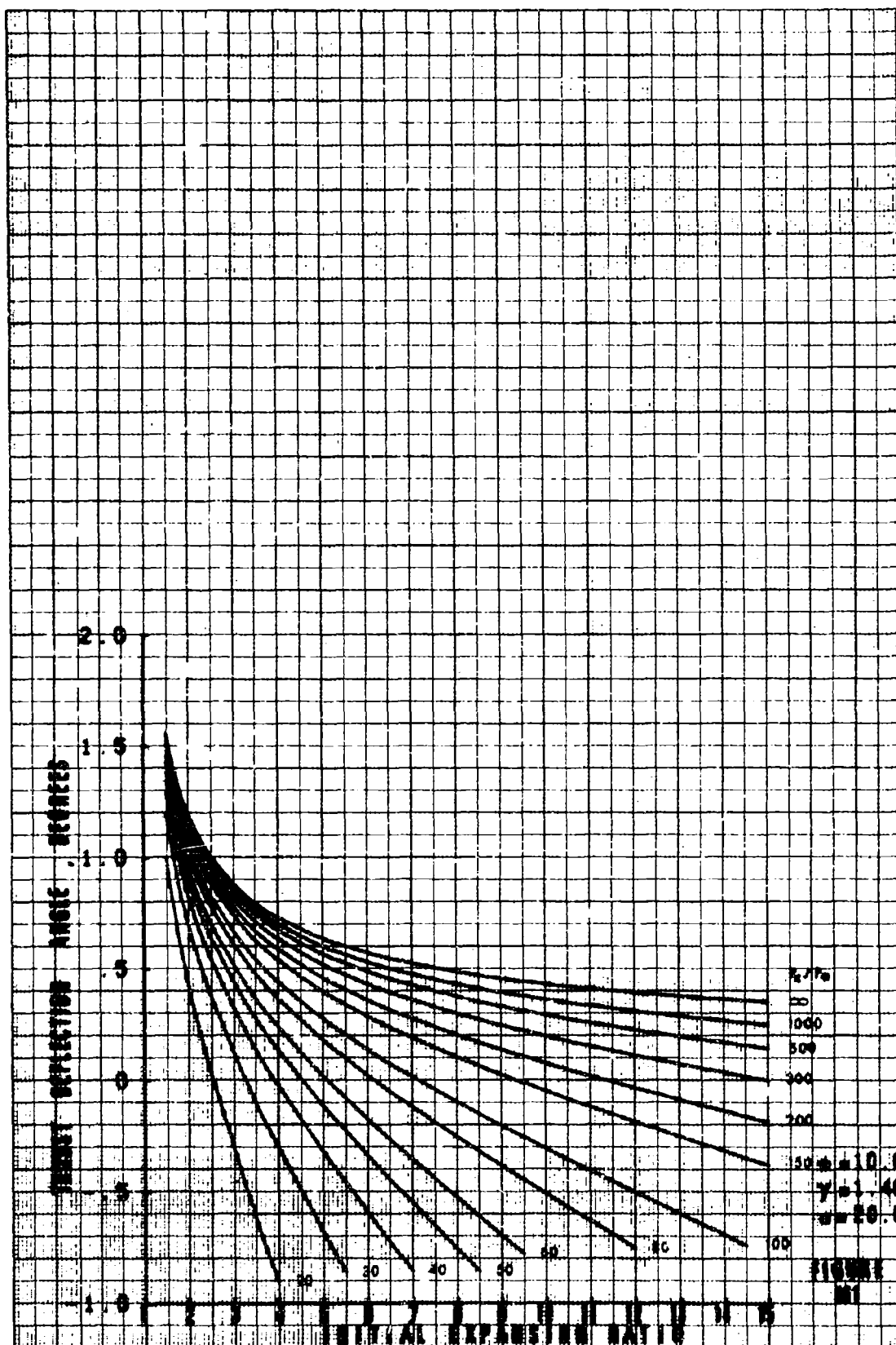


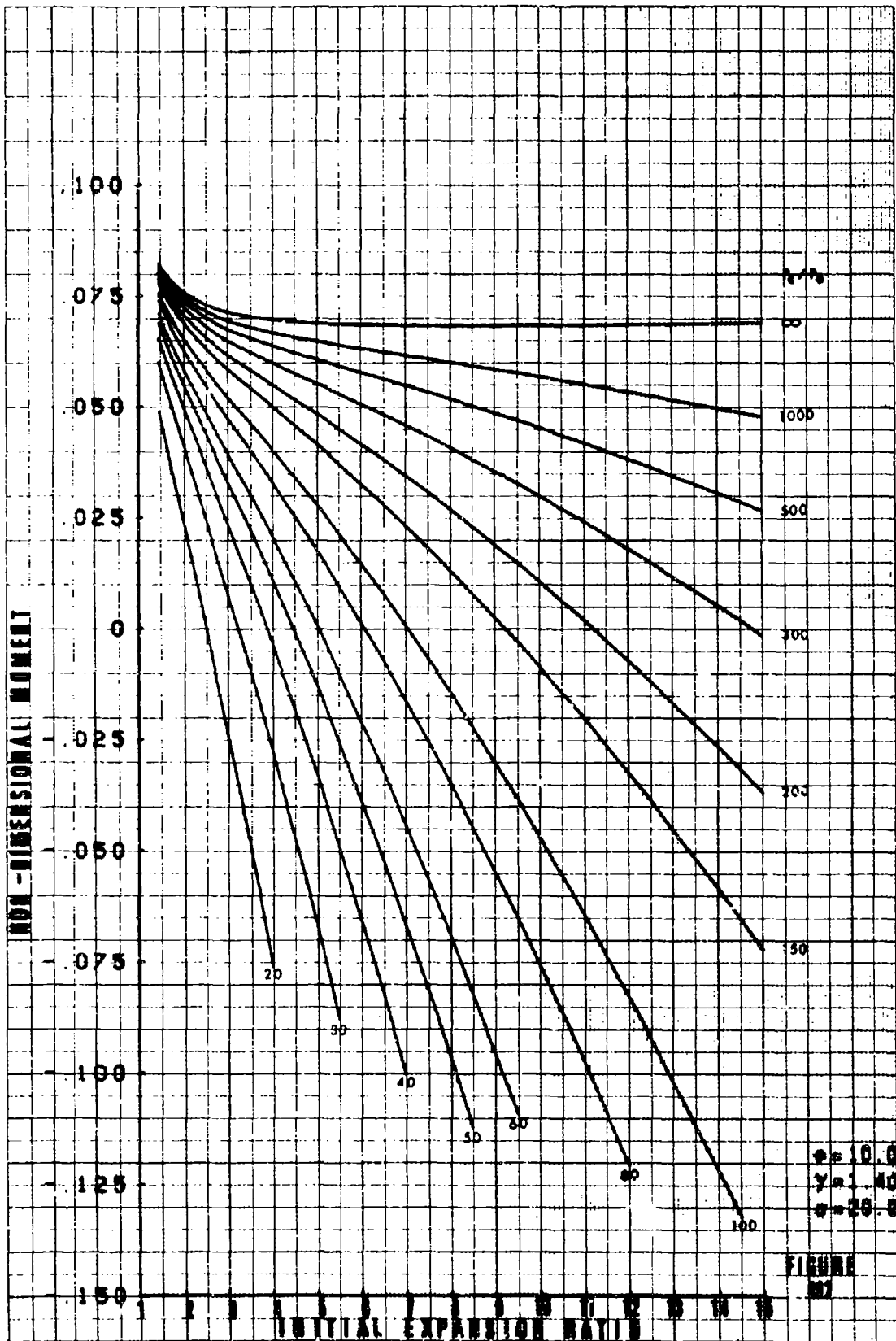












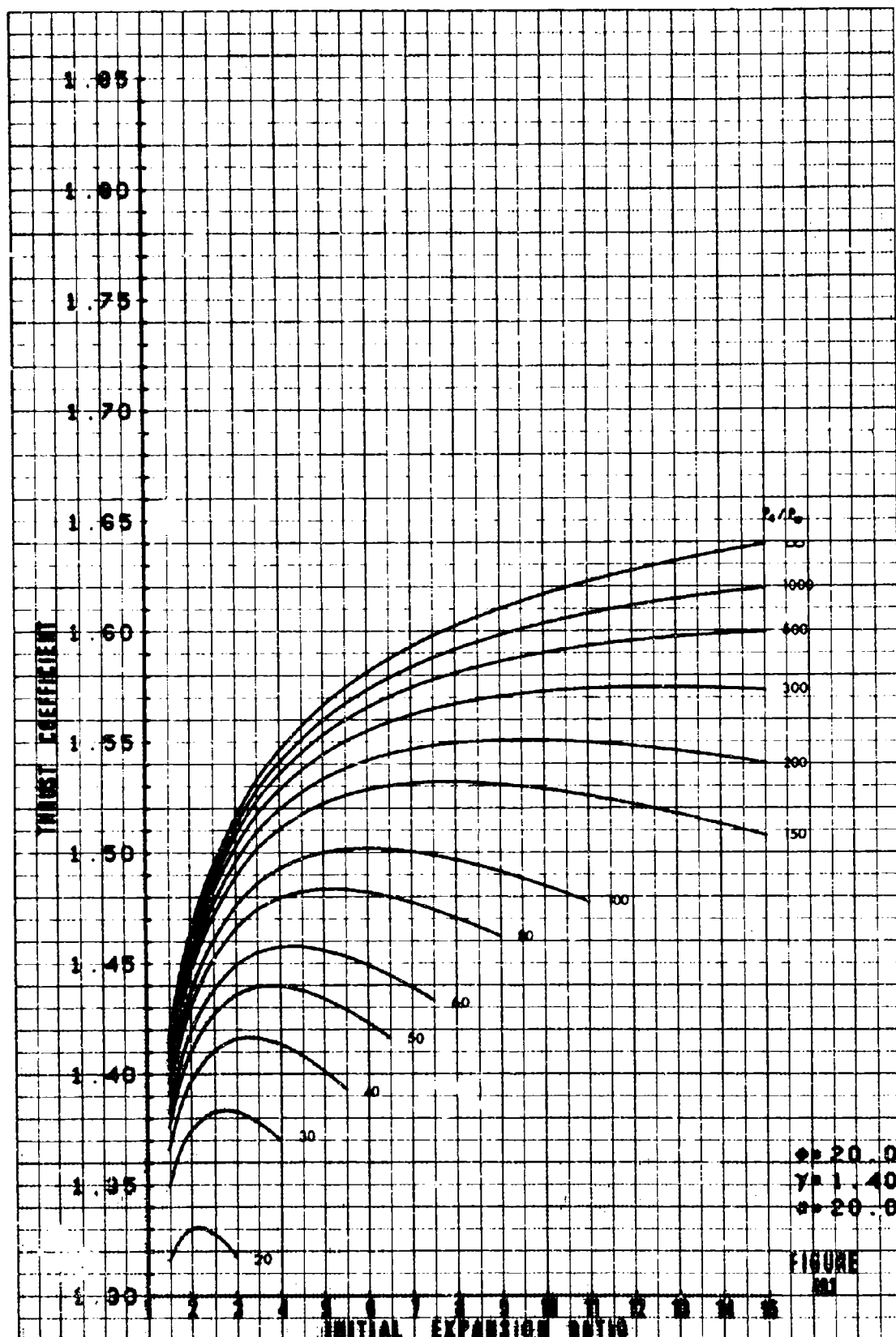
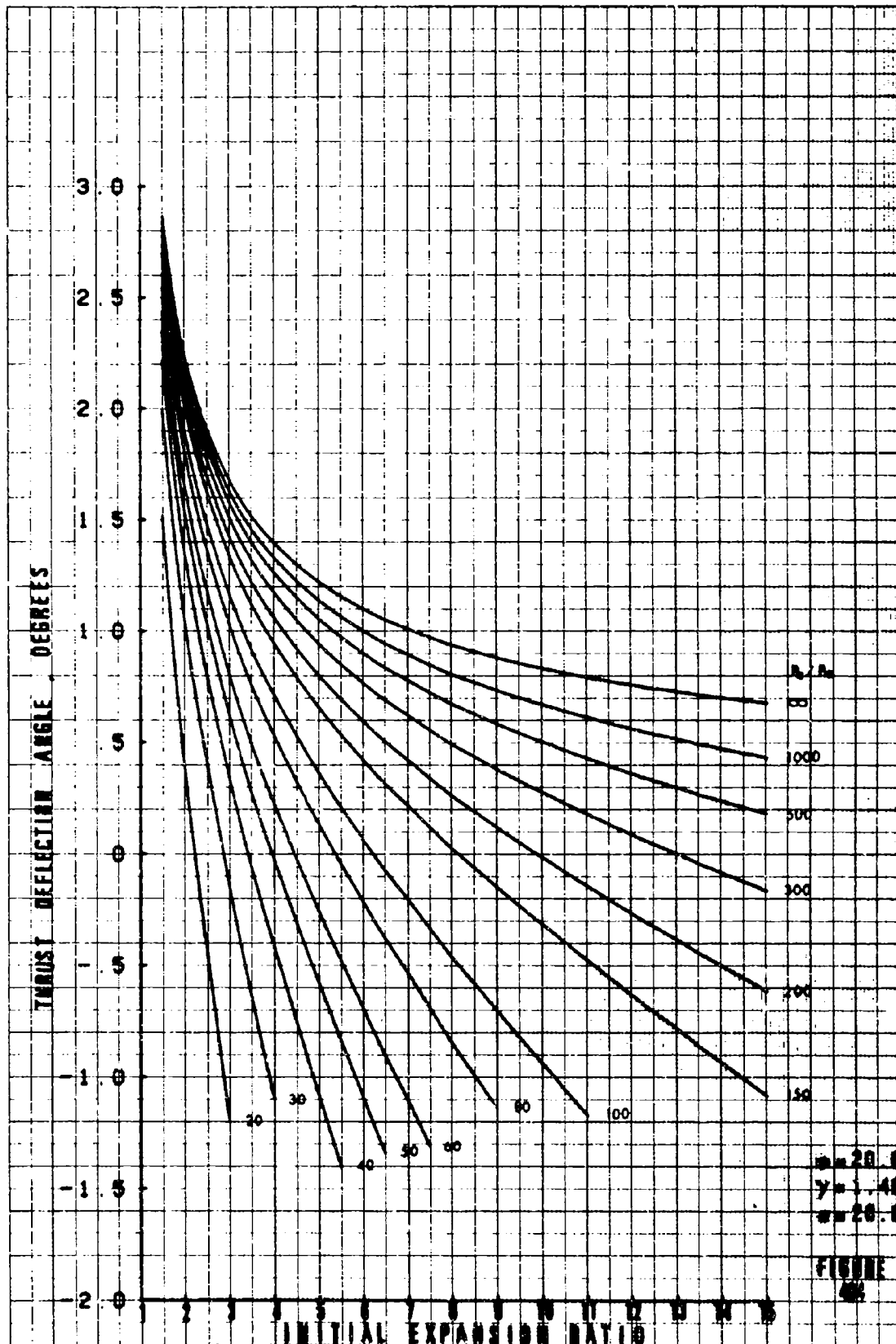
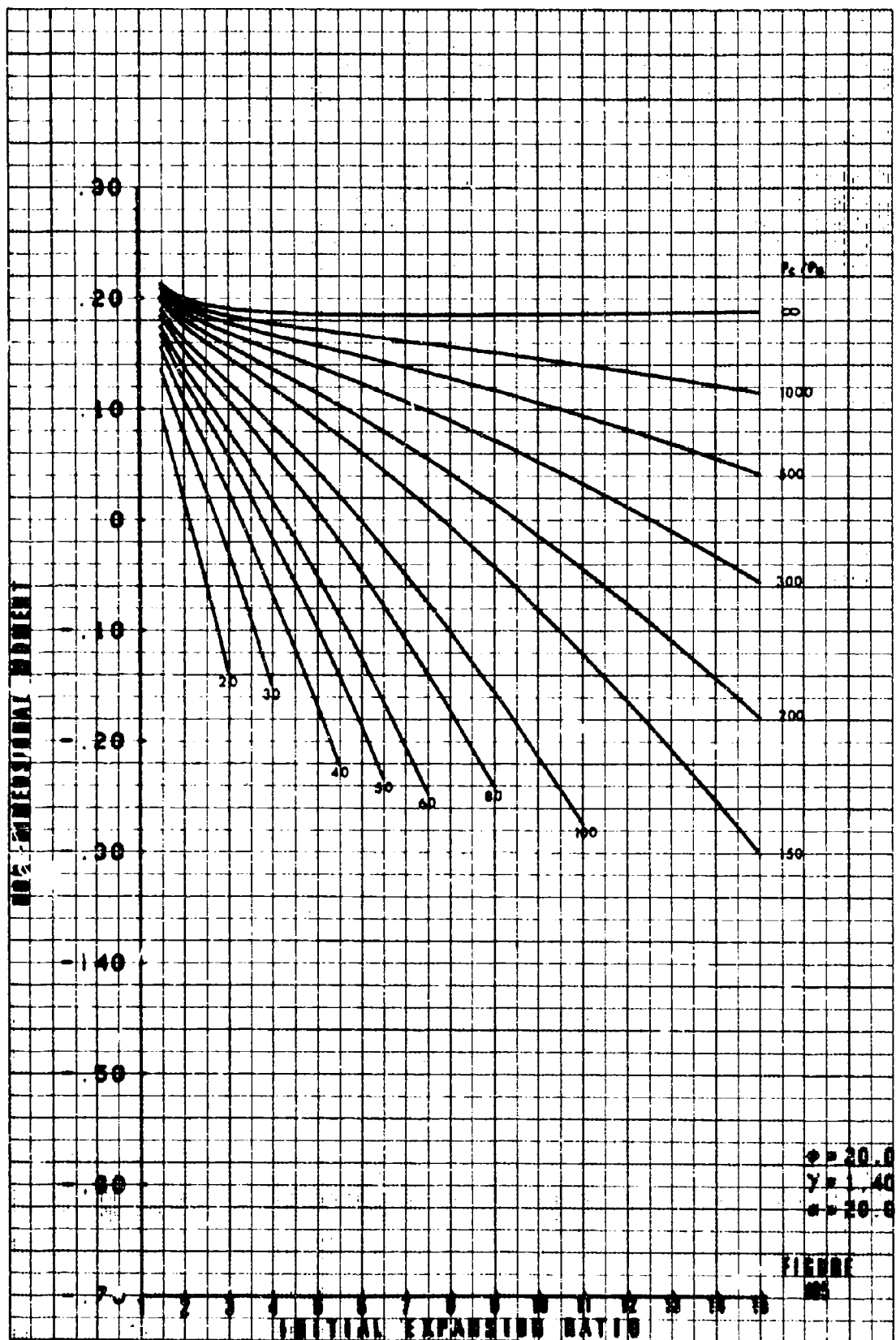
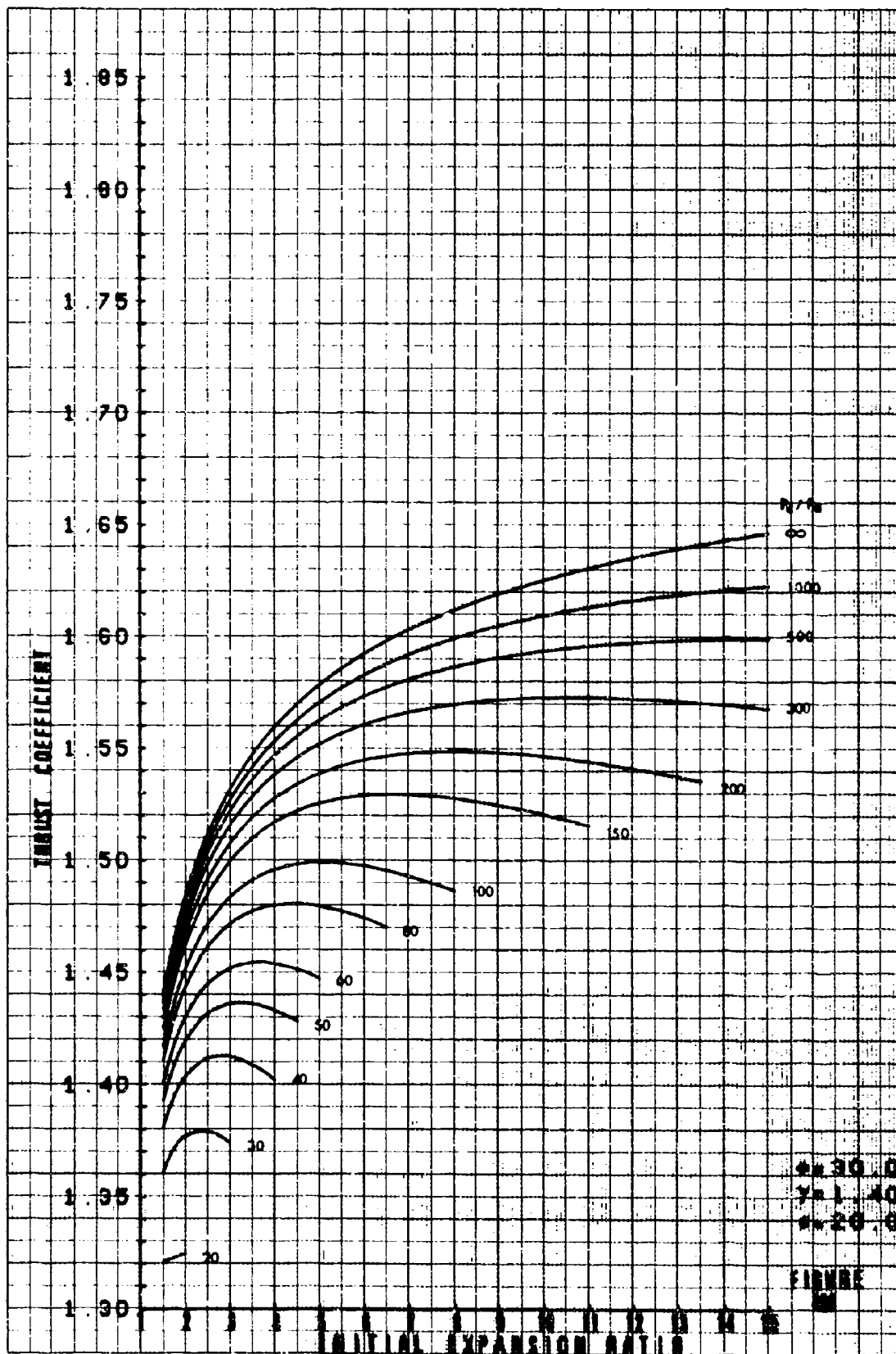


FIGURE 81

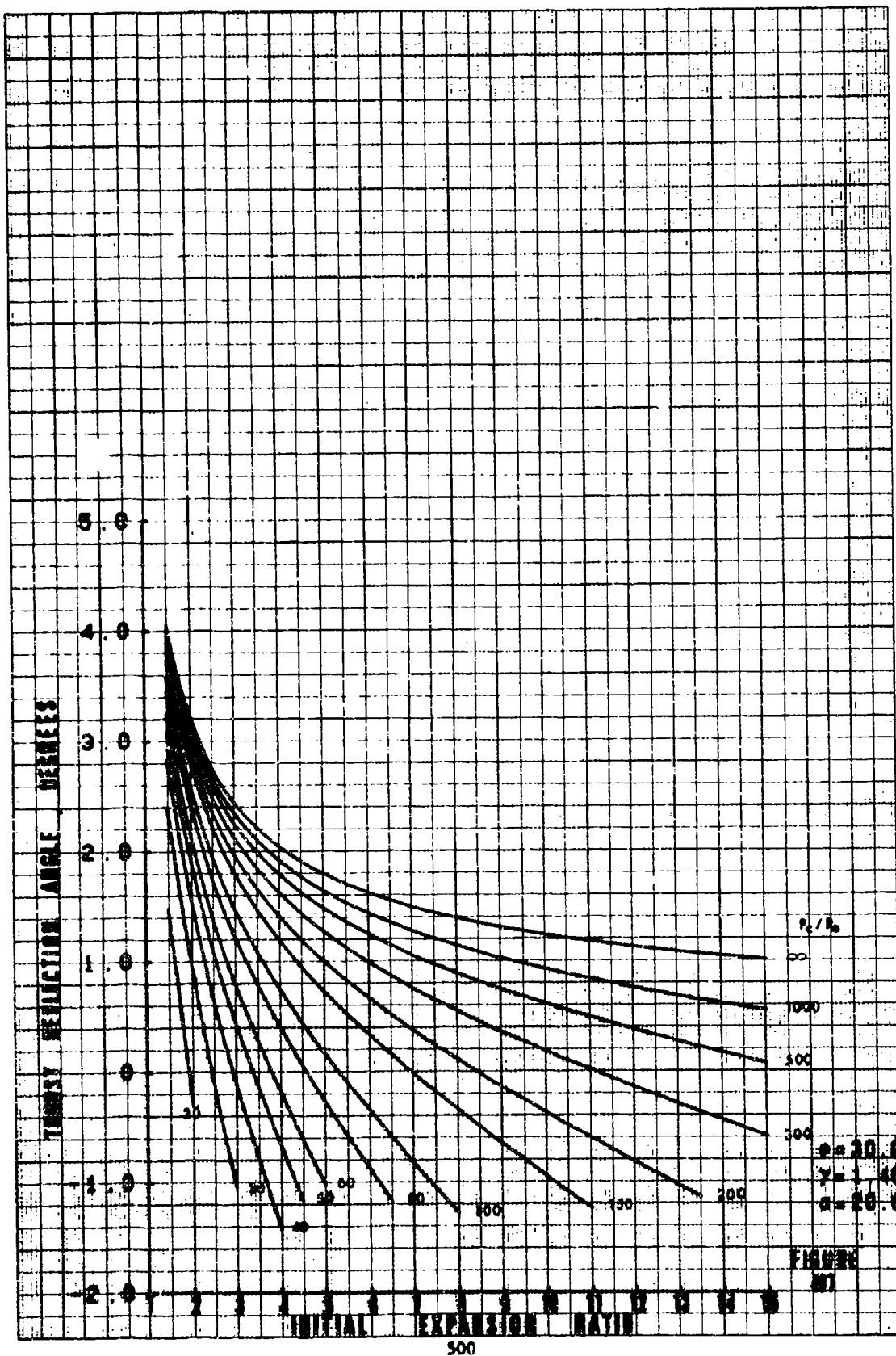


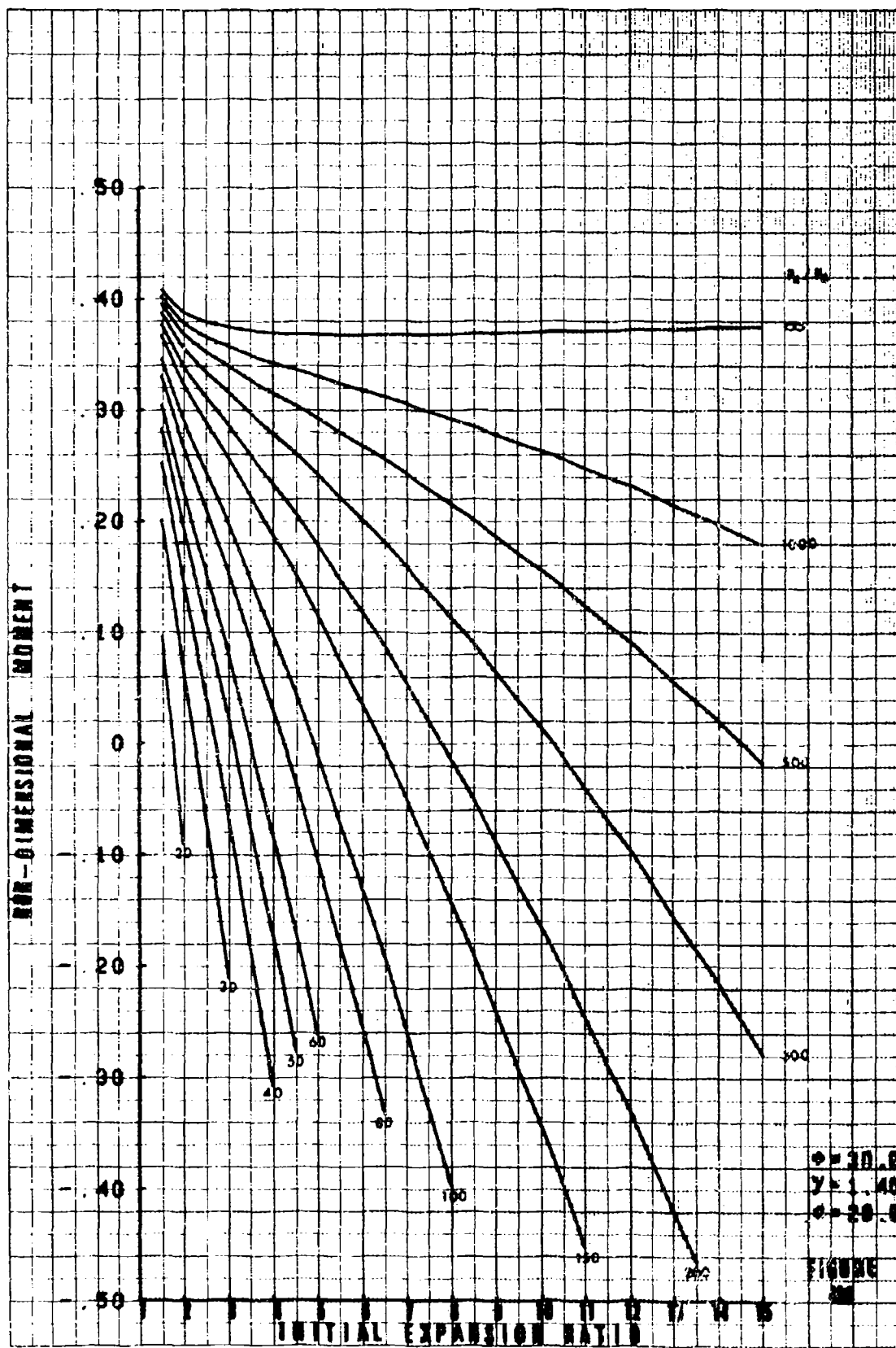


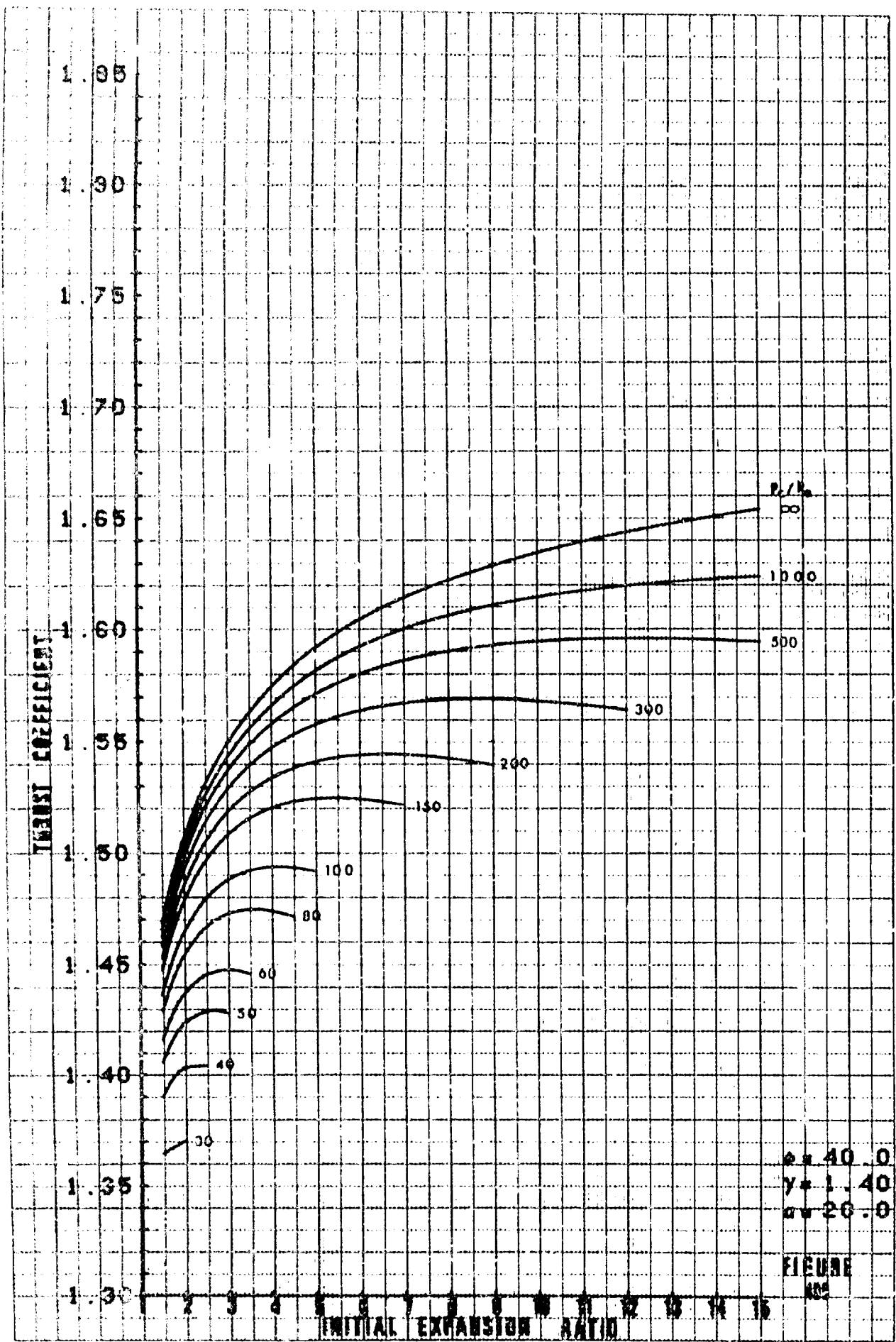


$\gamma = 1.40$
 $\gamma = 1.40$
 $\gamma = 1.40$

FIGURE







THRUST DEFLECTION ANGLE, DEGREES

6.0
5.0
4.0
3.0
2.0
1.0
0
-1.0

30
40
50
60
80
100
150
200
300

INITIAL

EXPANSION

RATIO

P_t / P_0

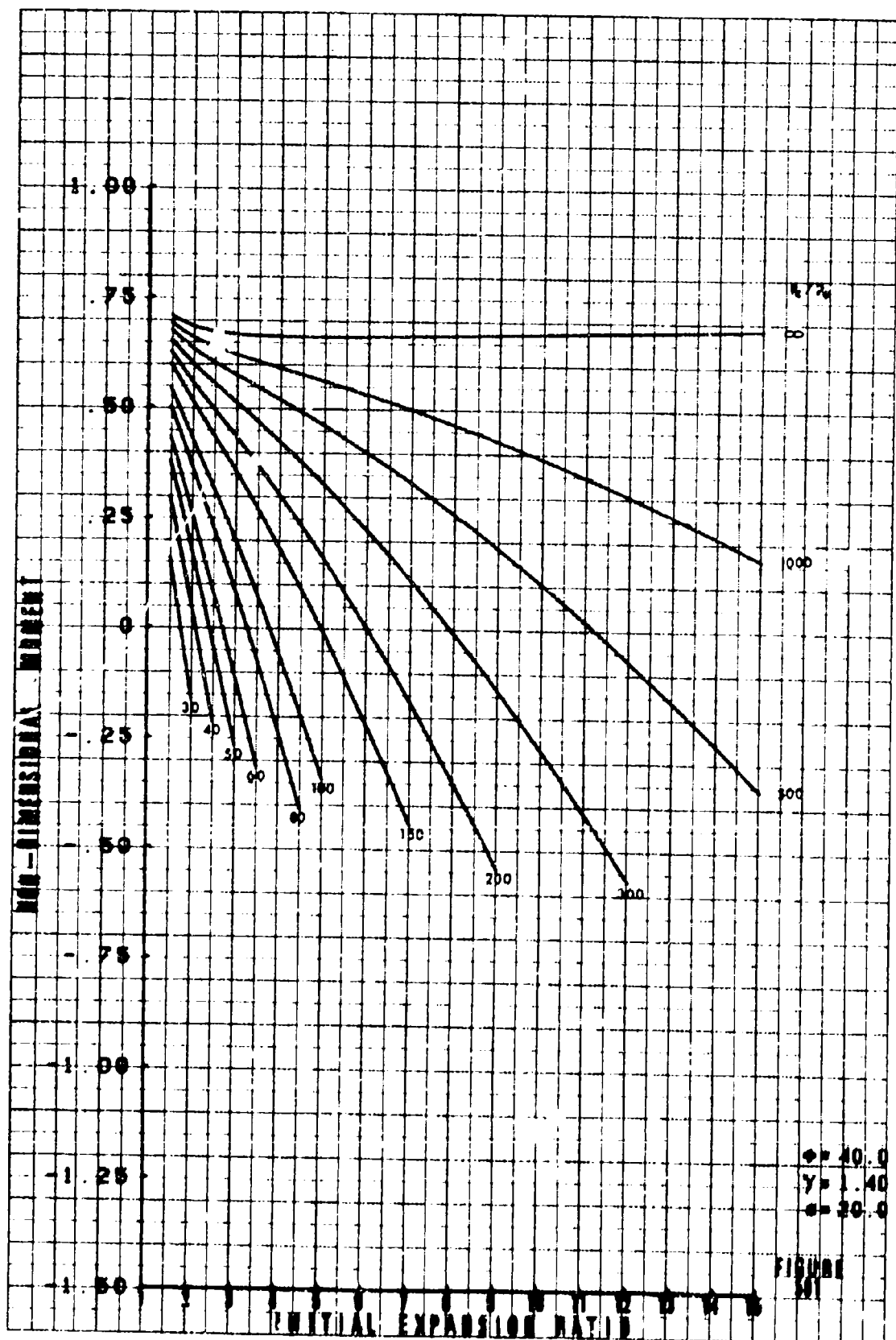
1000

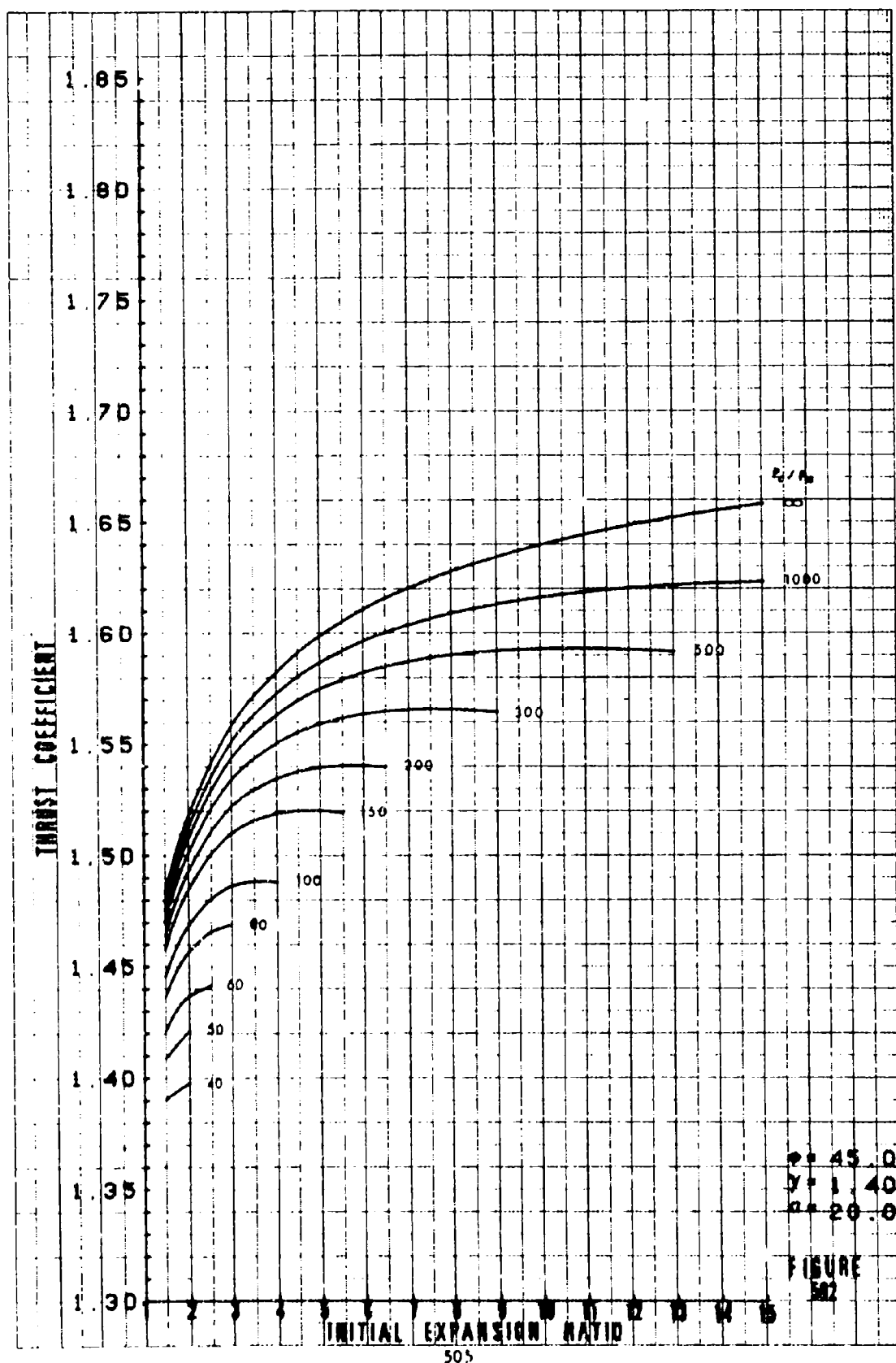
$\phi = 40.0$

$\gamma = 1.40$

$\alpha = 20.6$

FIGURE 503





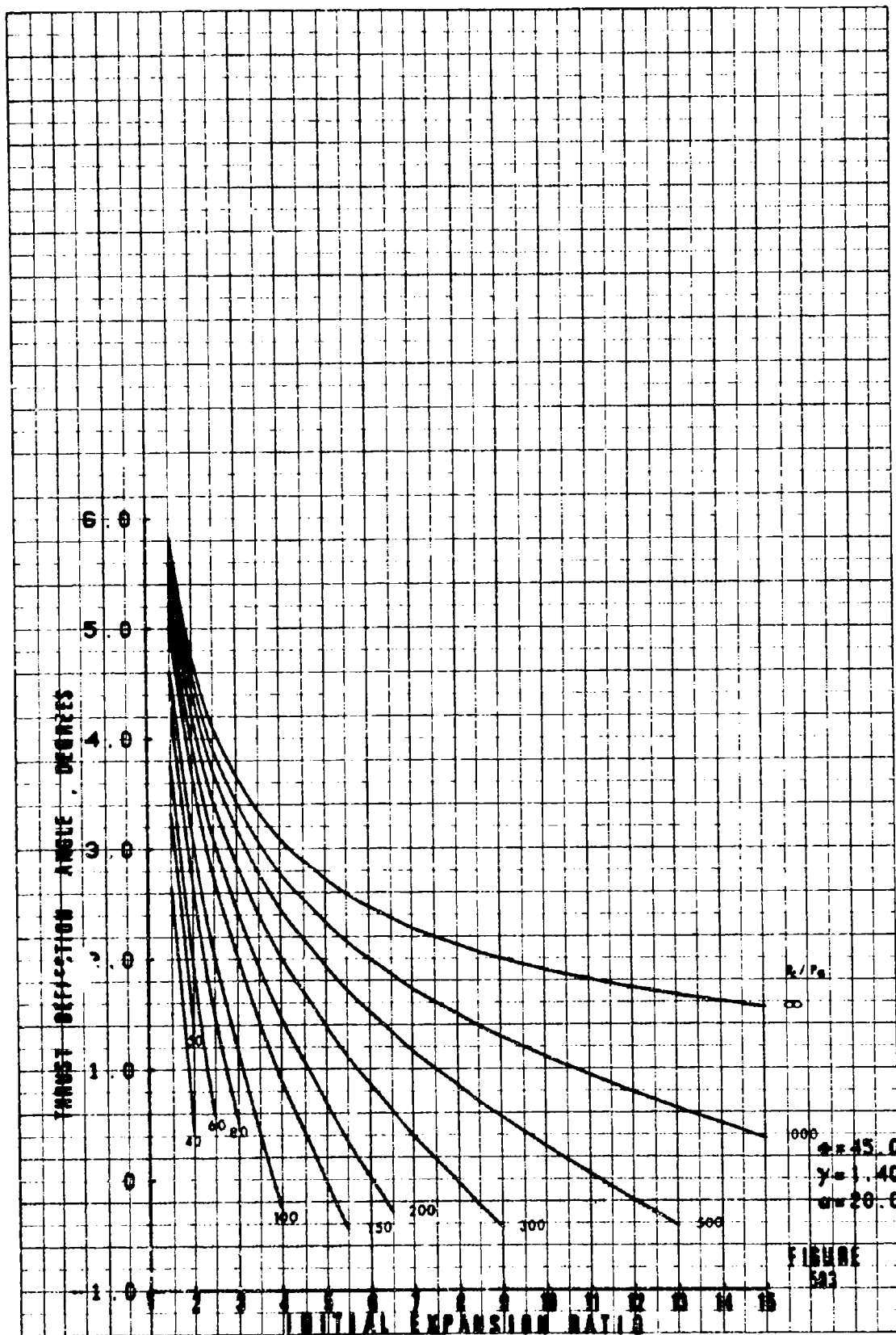
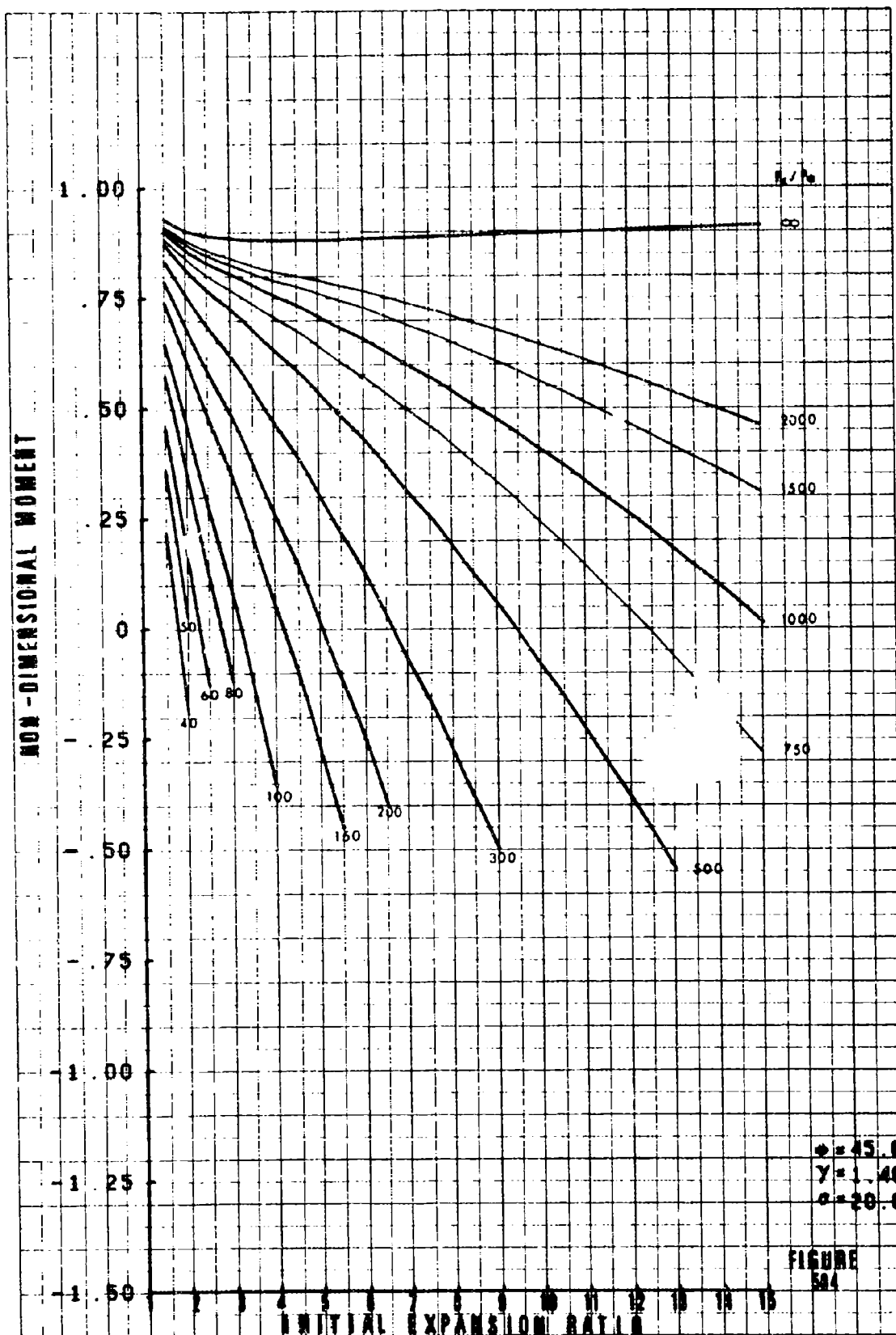
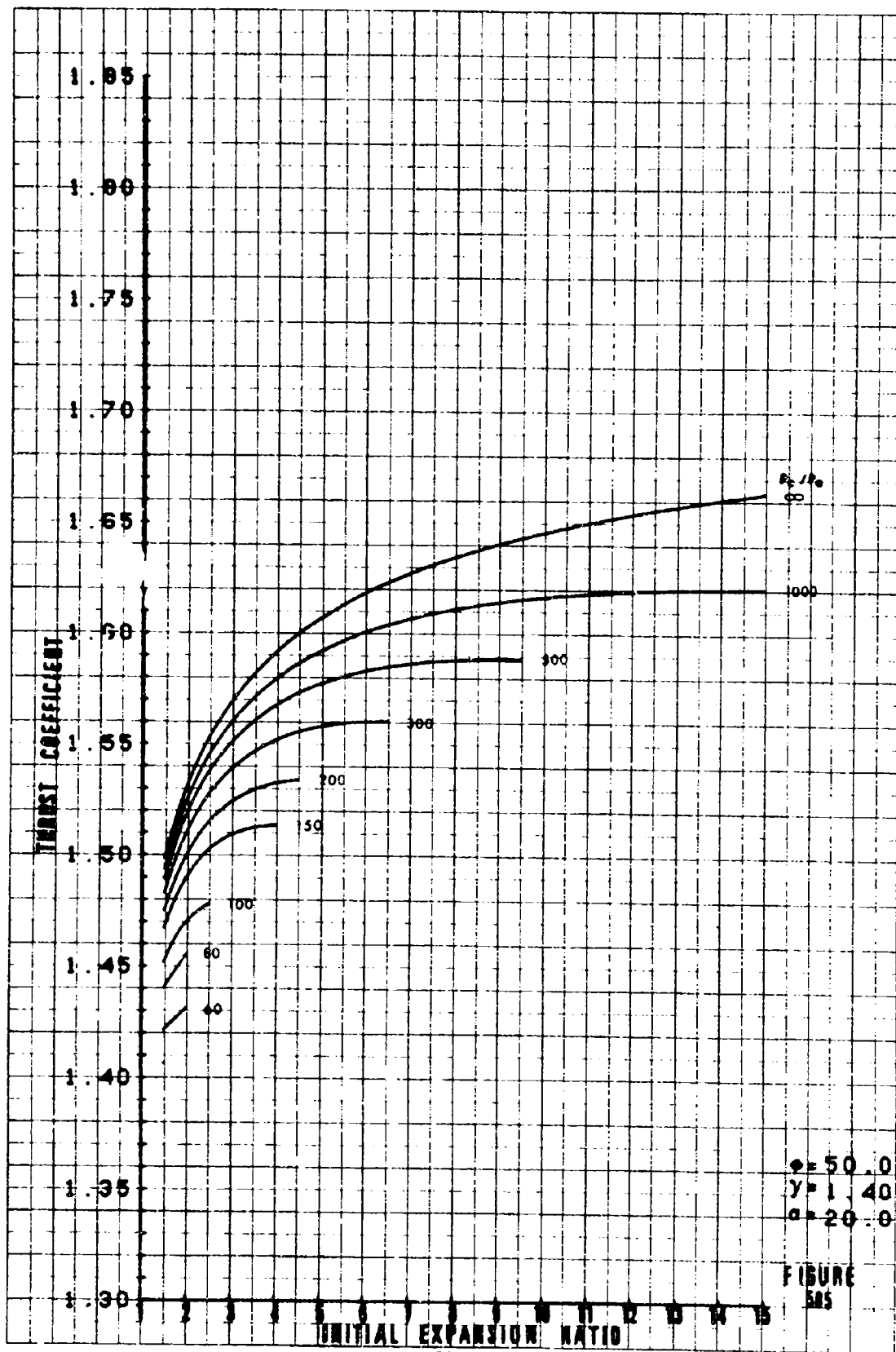


FIGURE 503





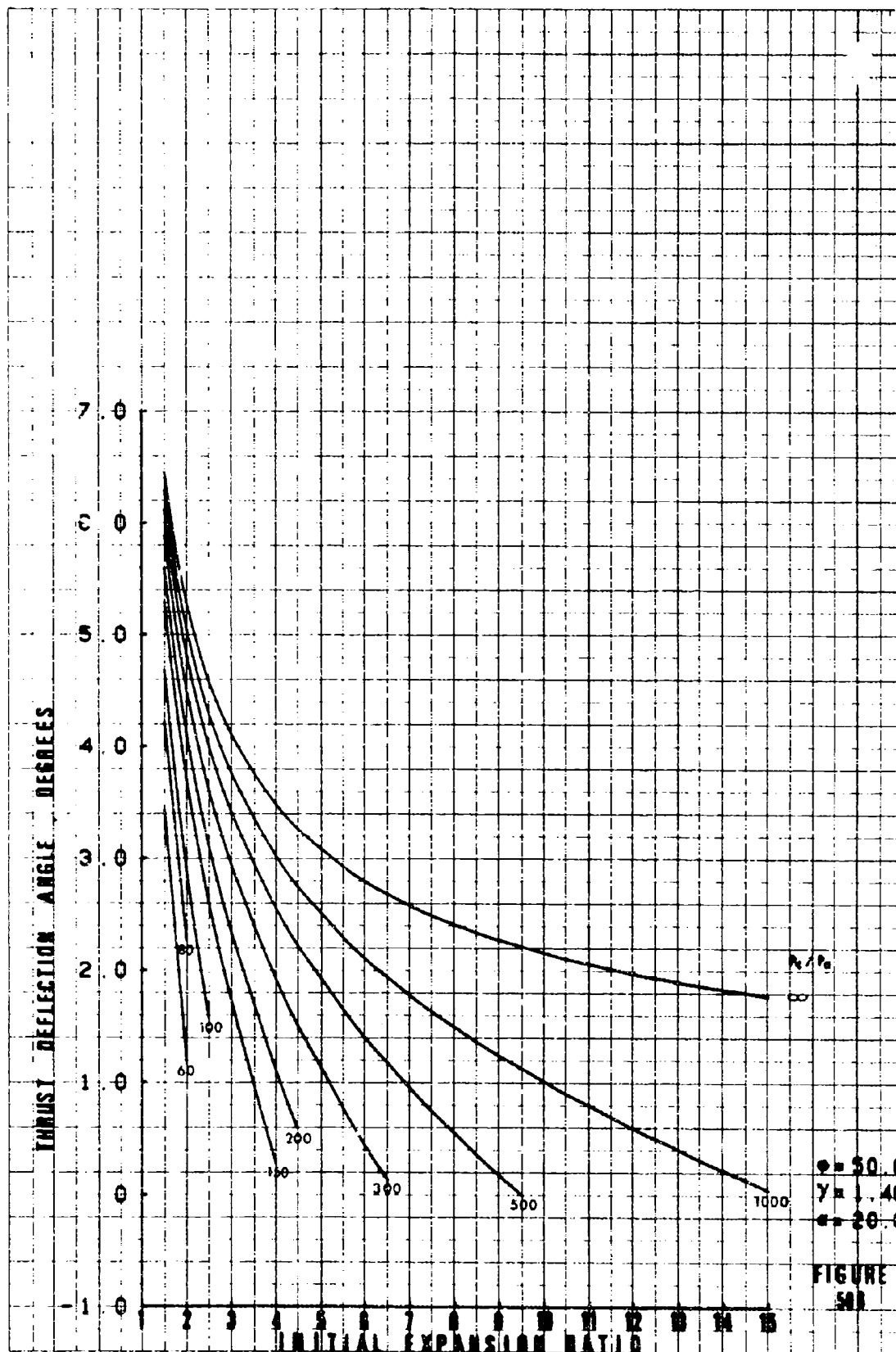
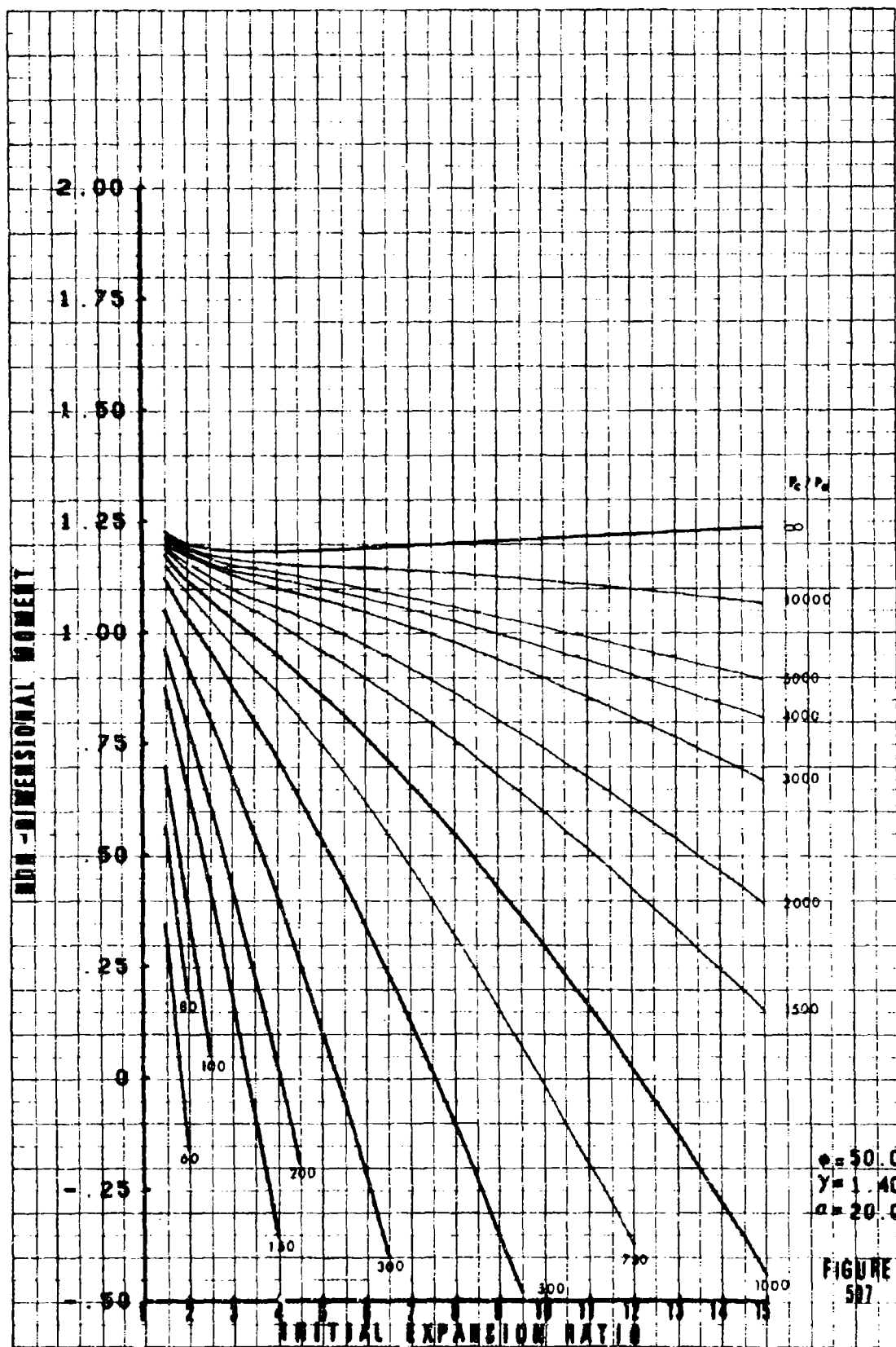
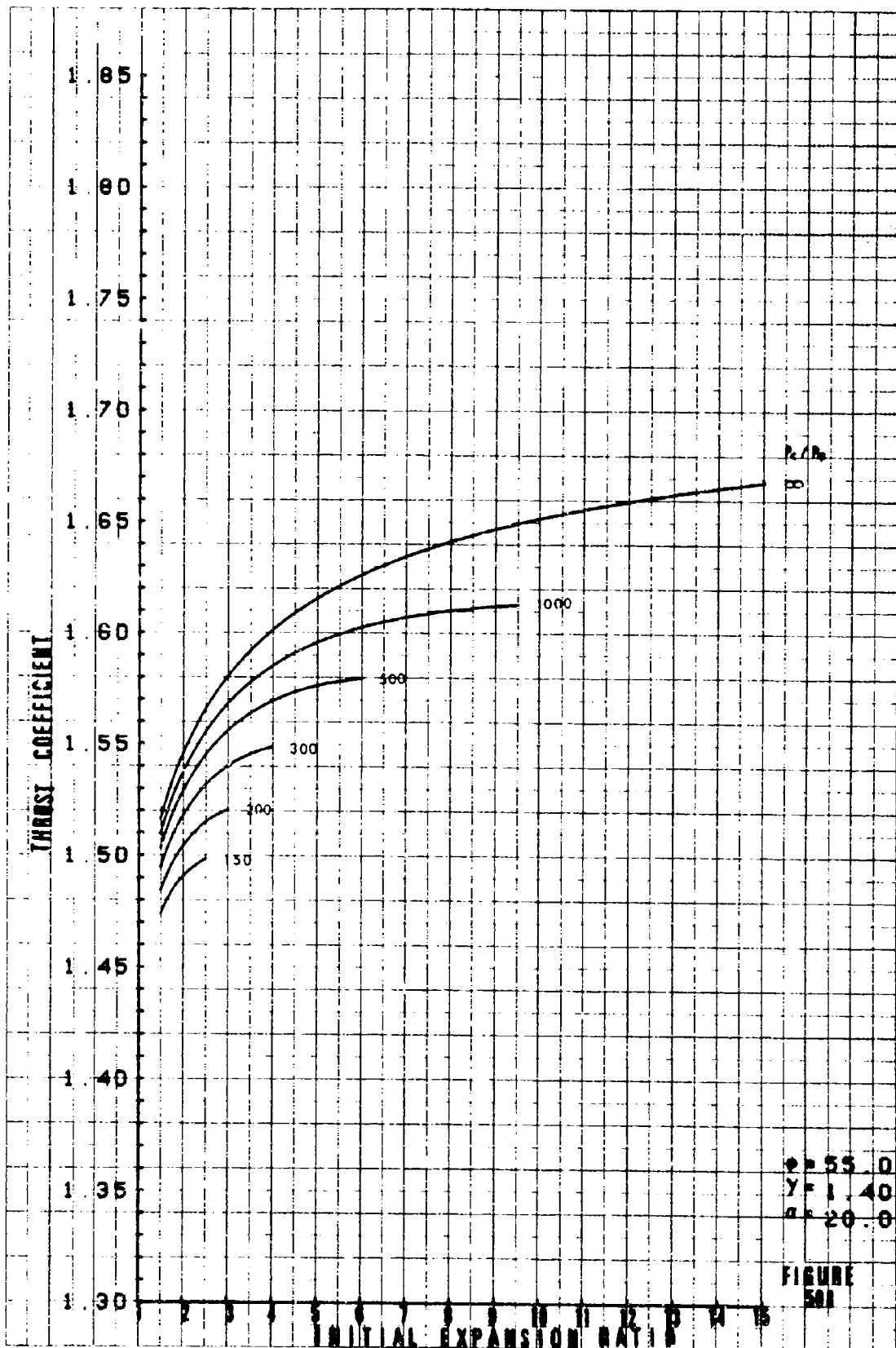


FIGURE 50





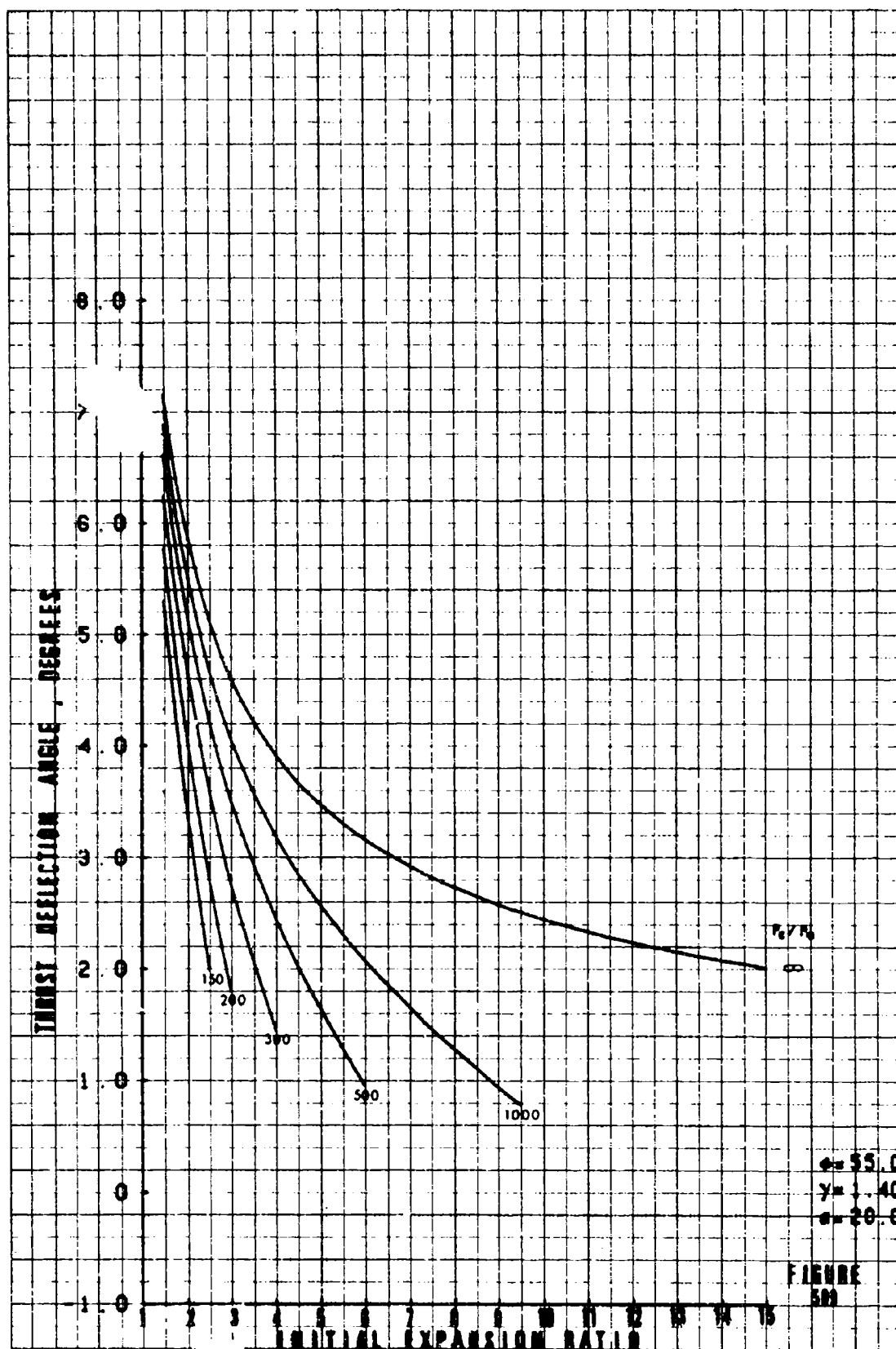
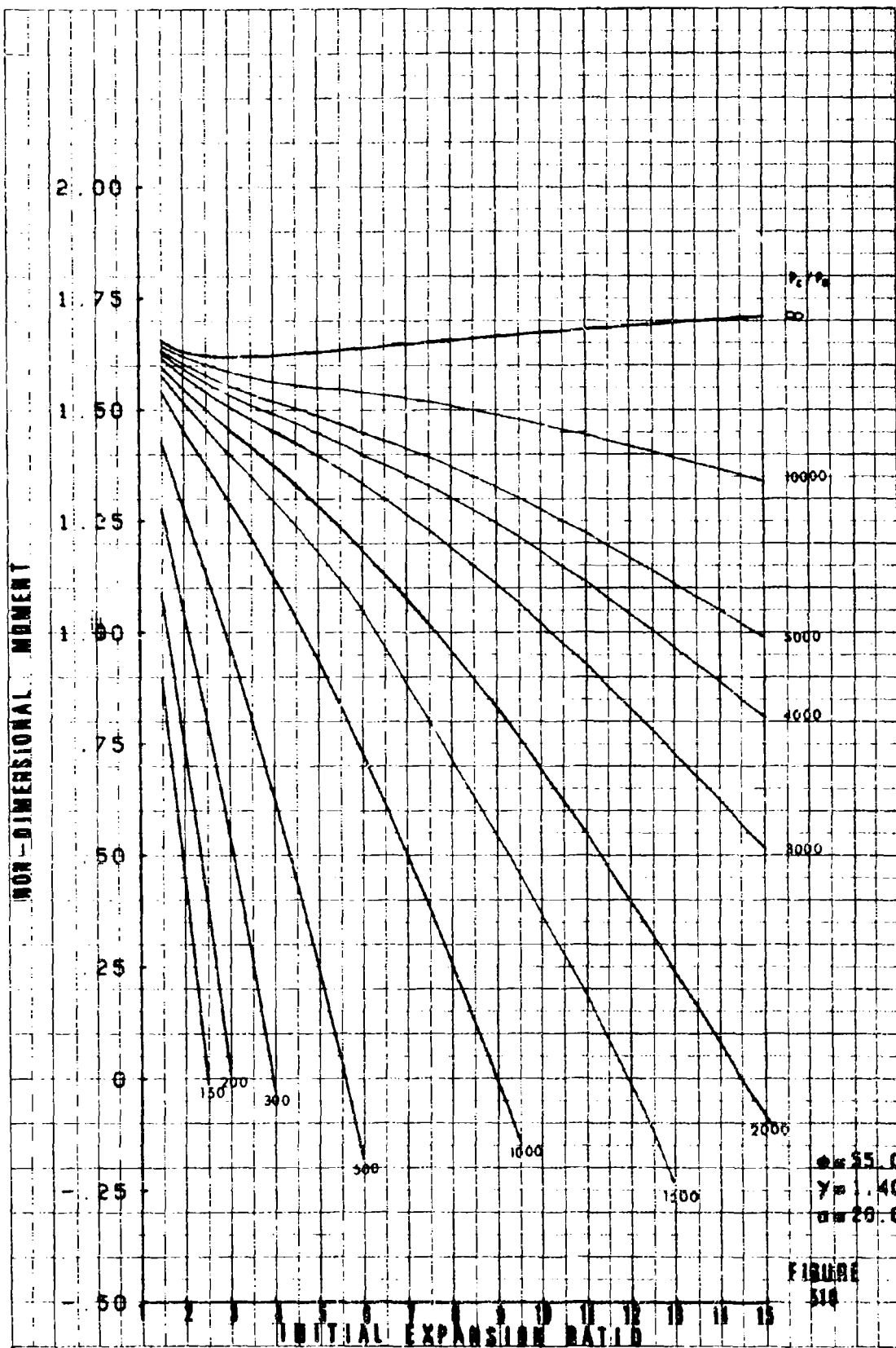
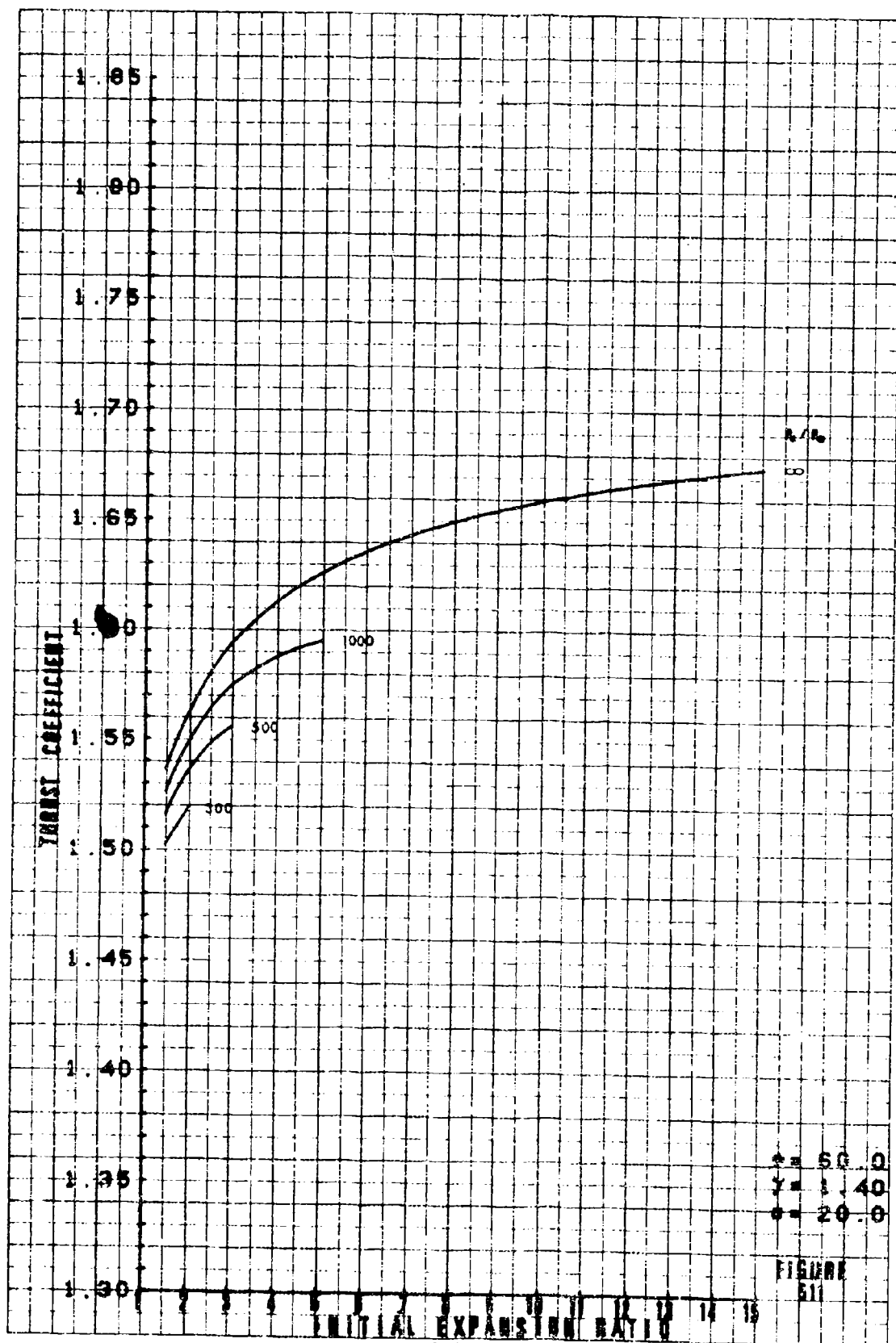
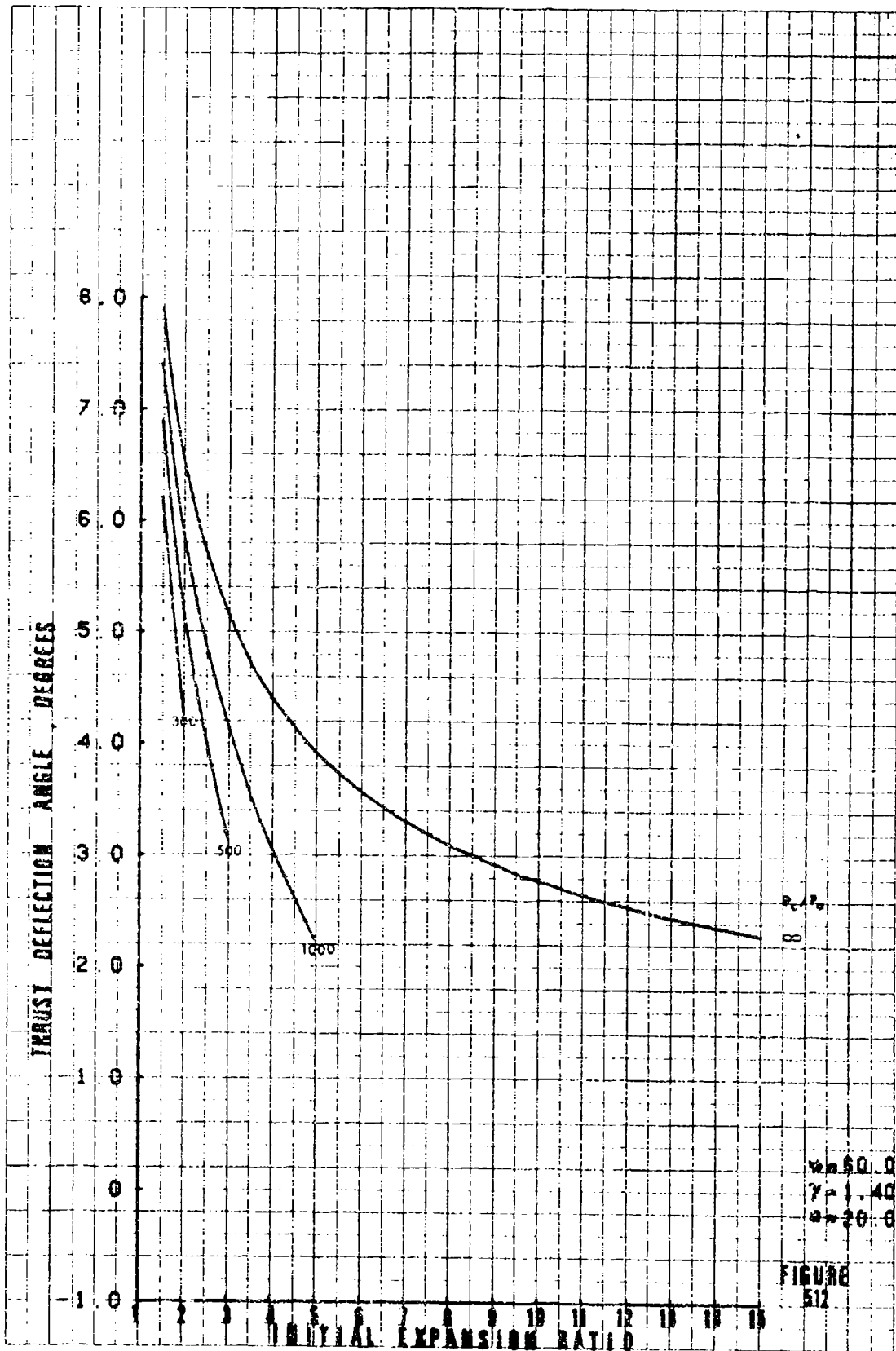
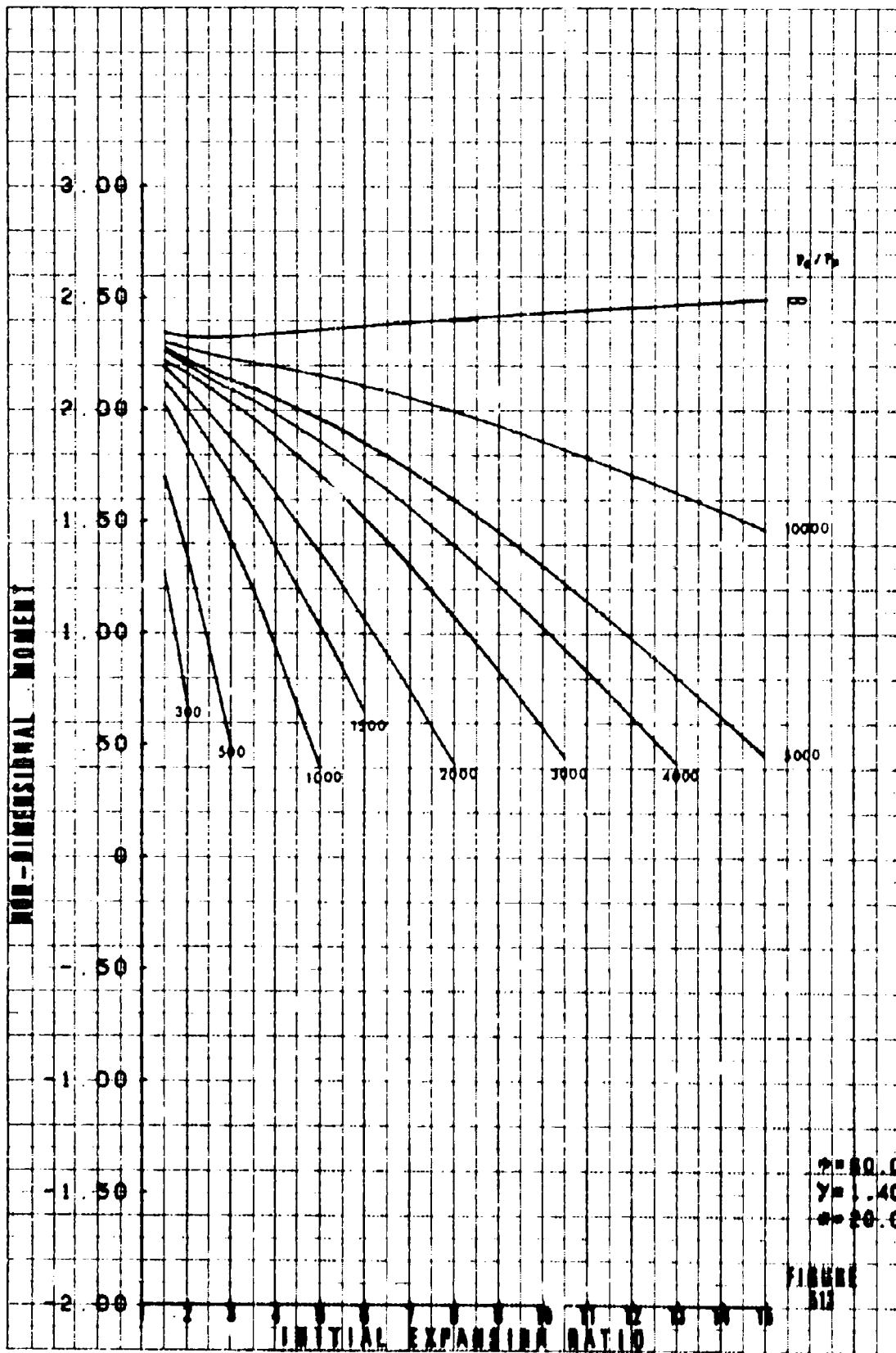


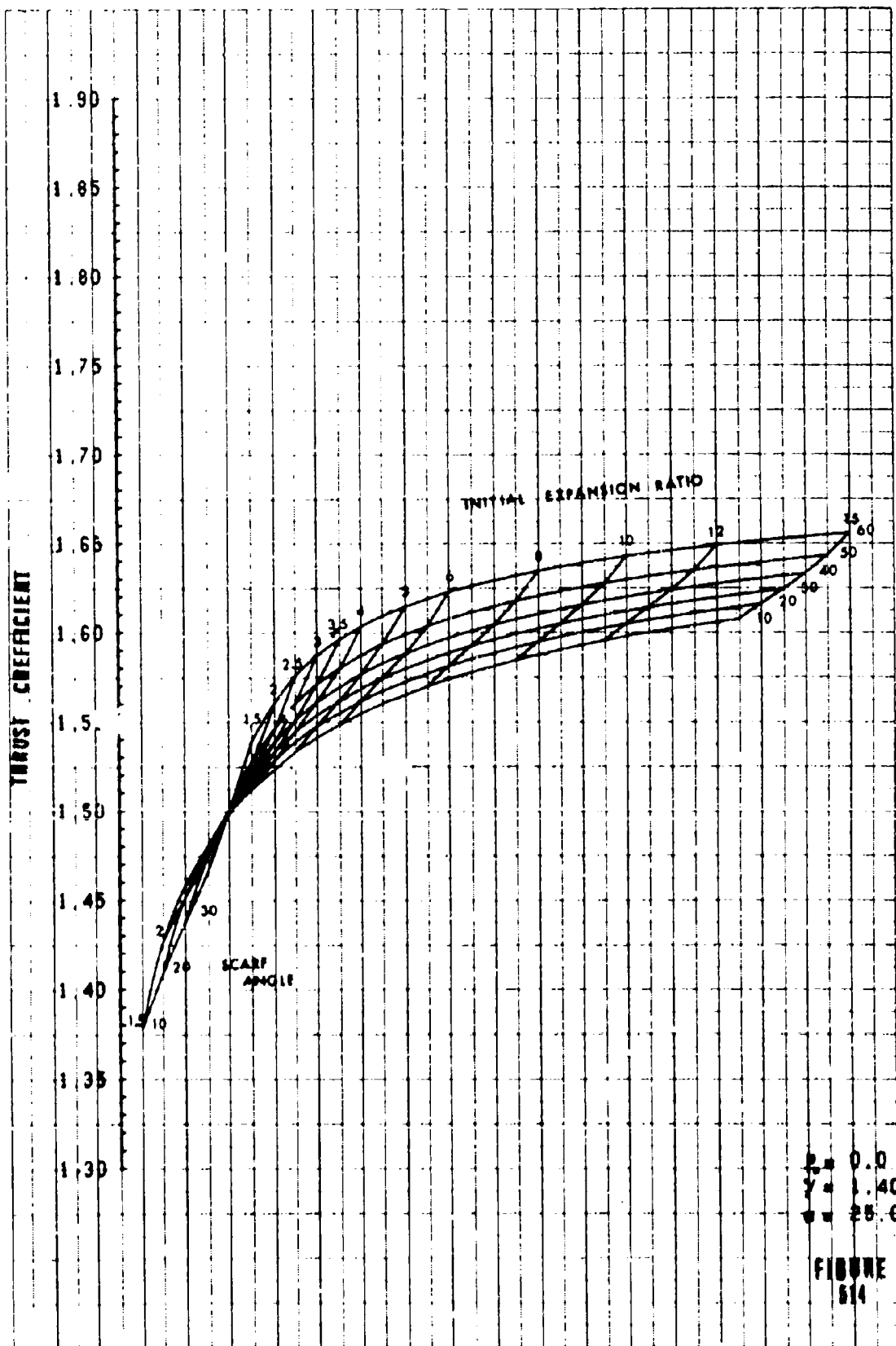
FIGURE 509

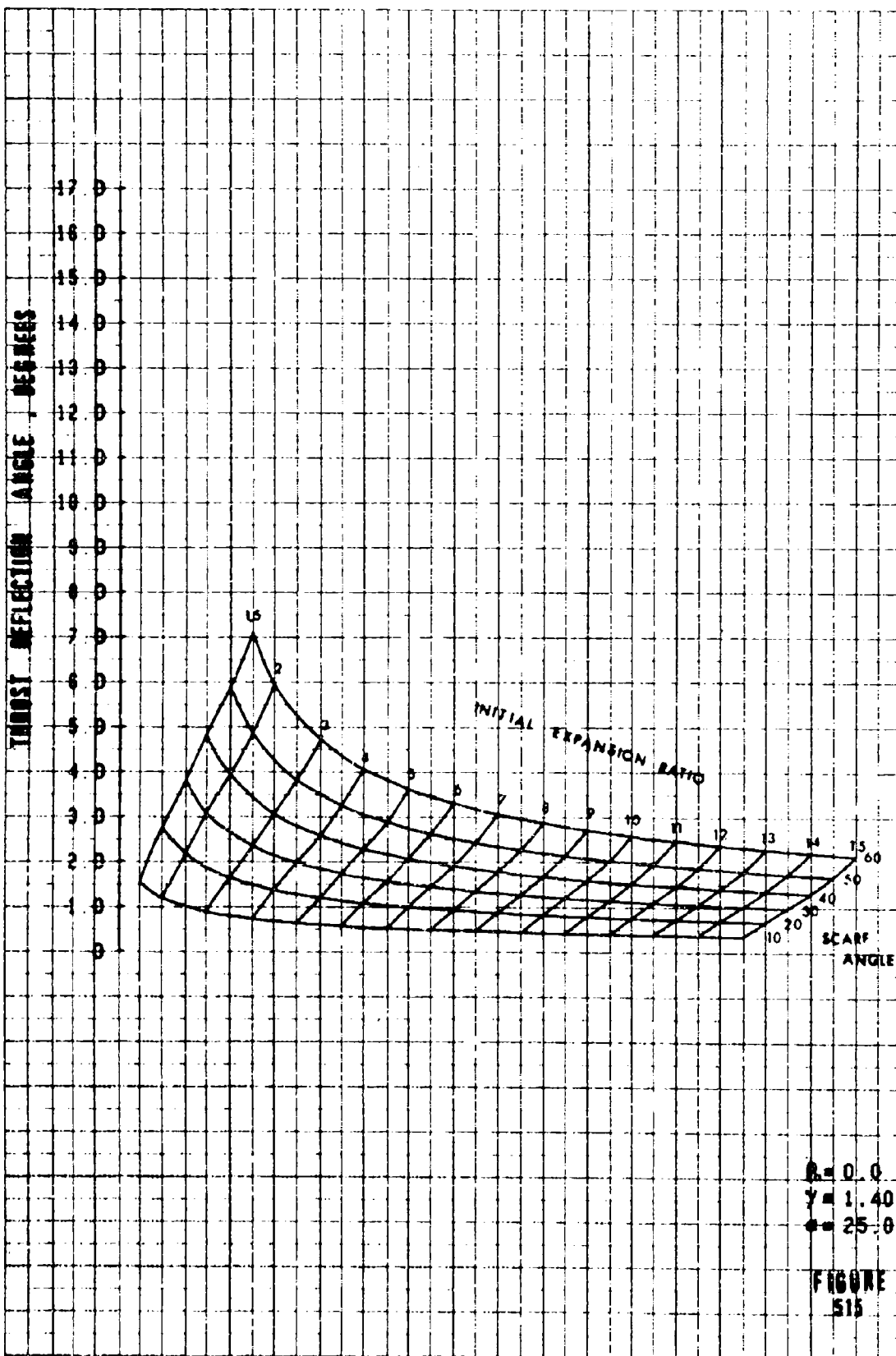


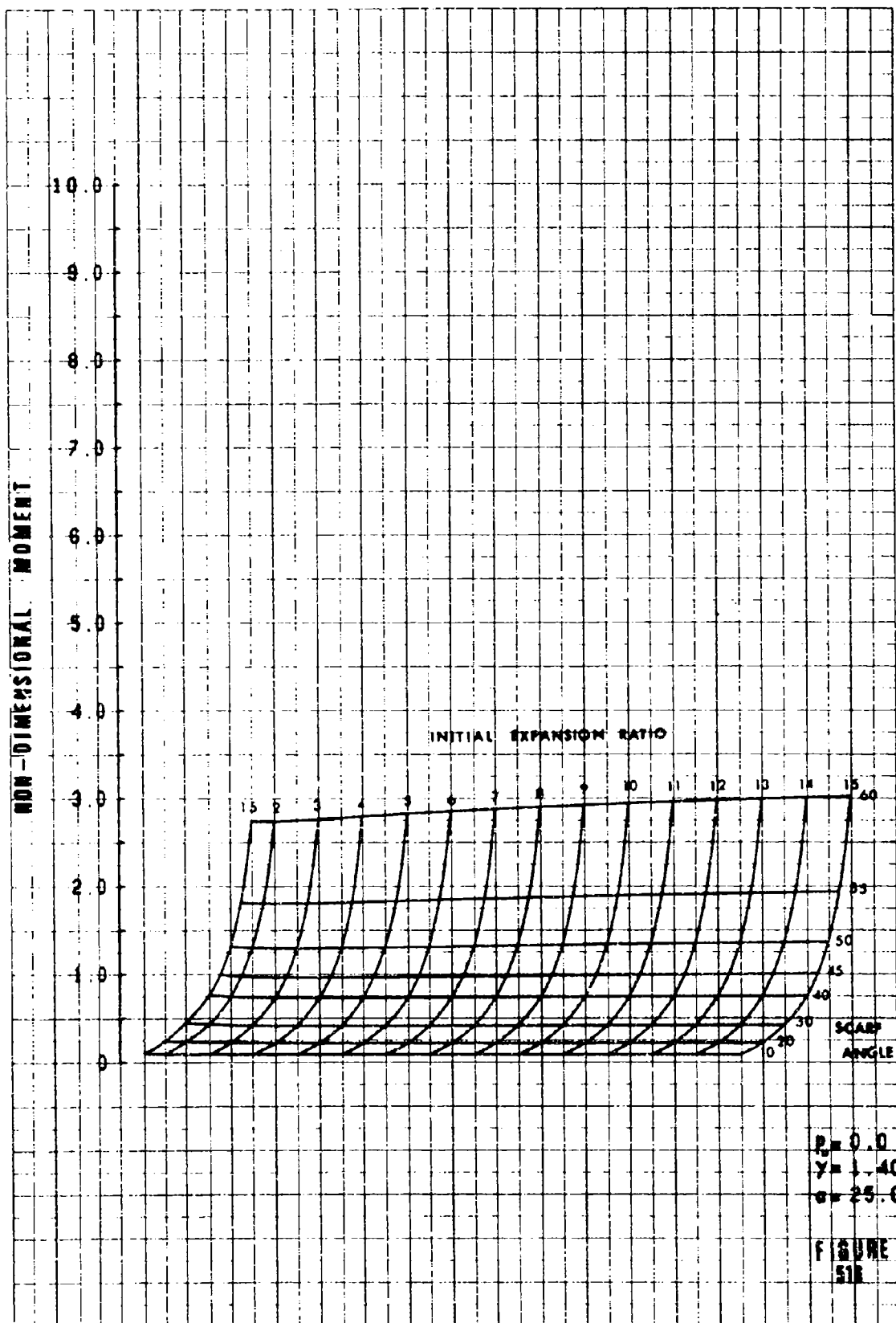












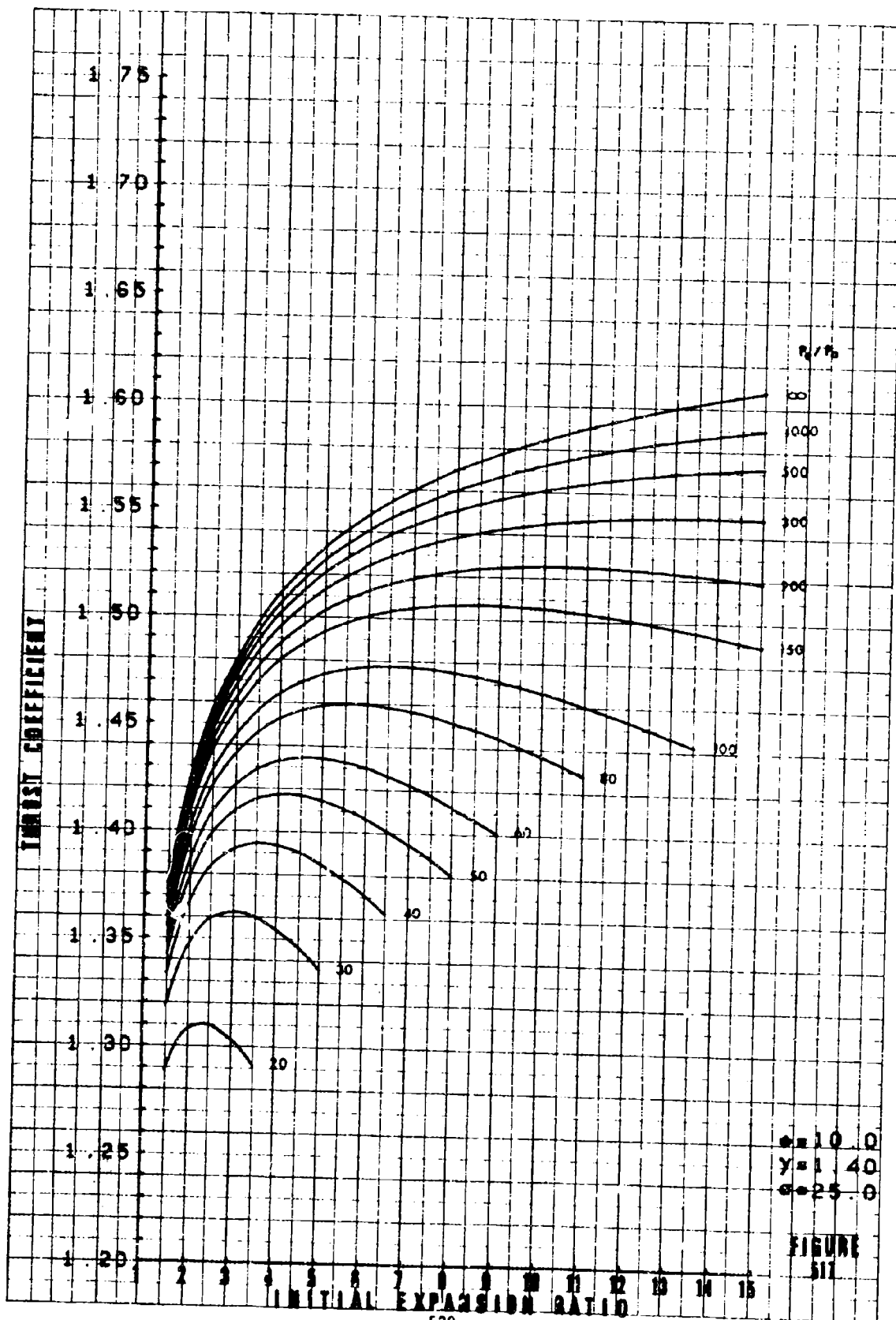
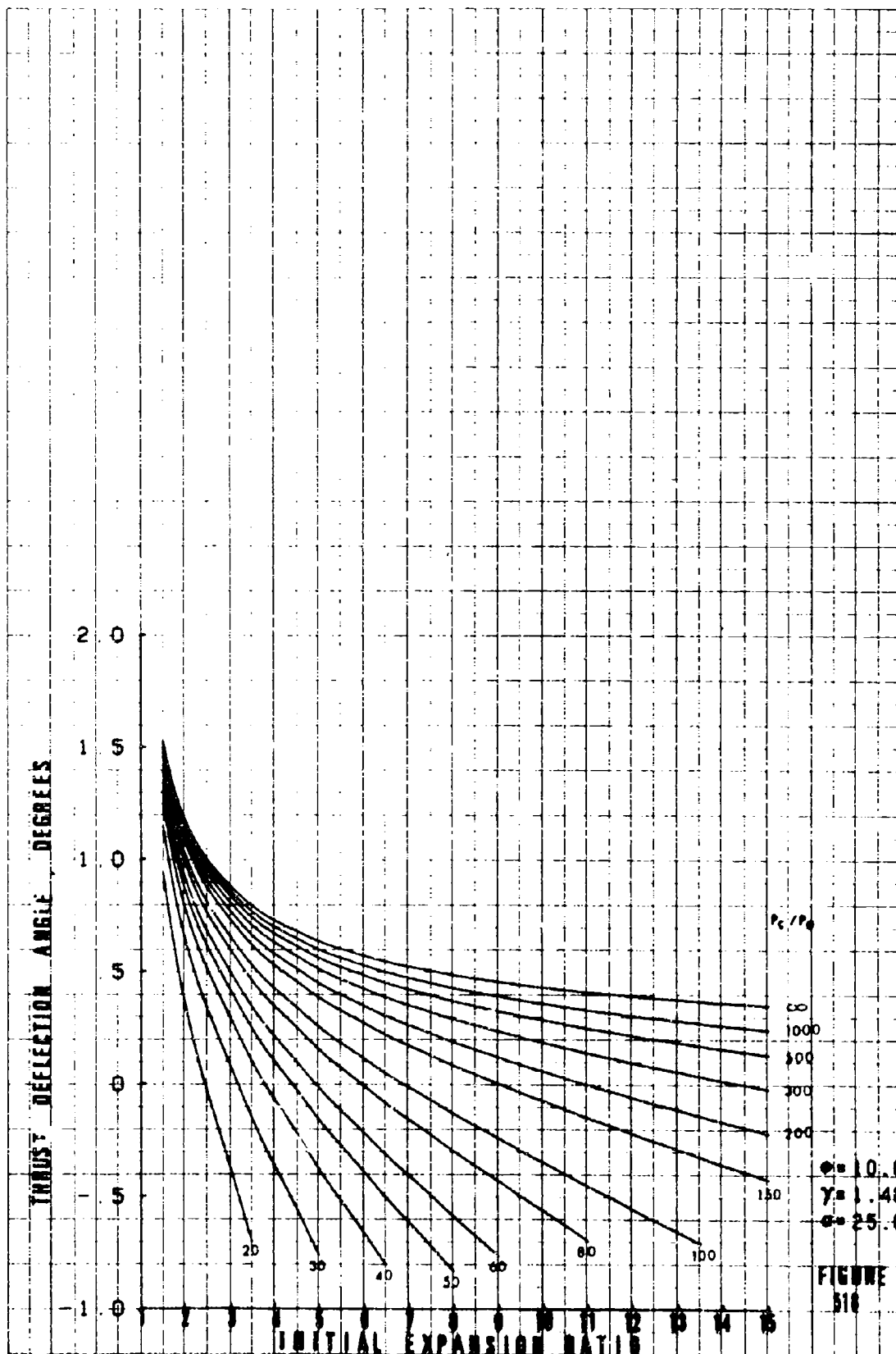
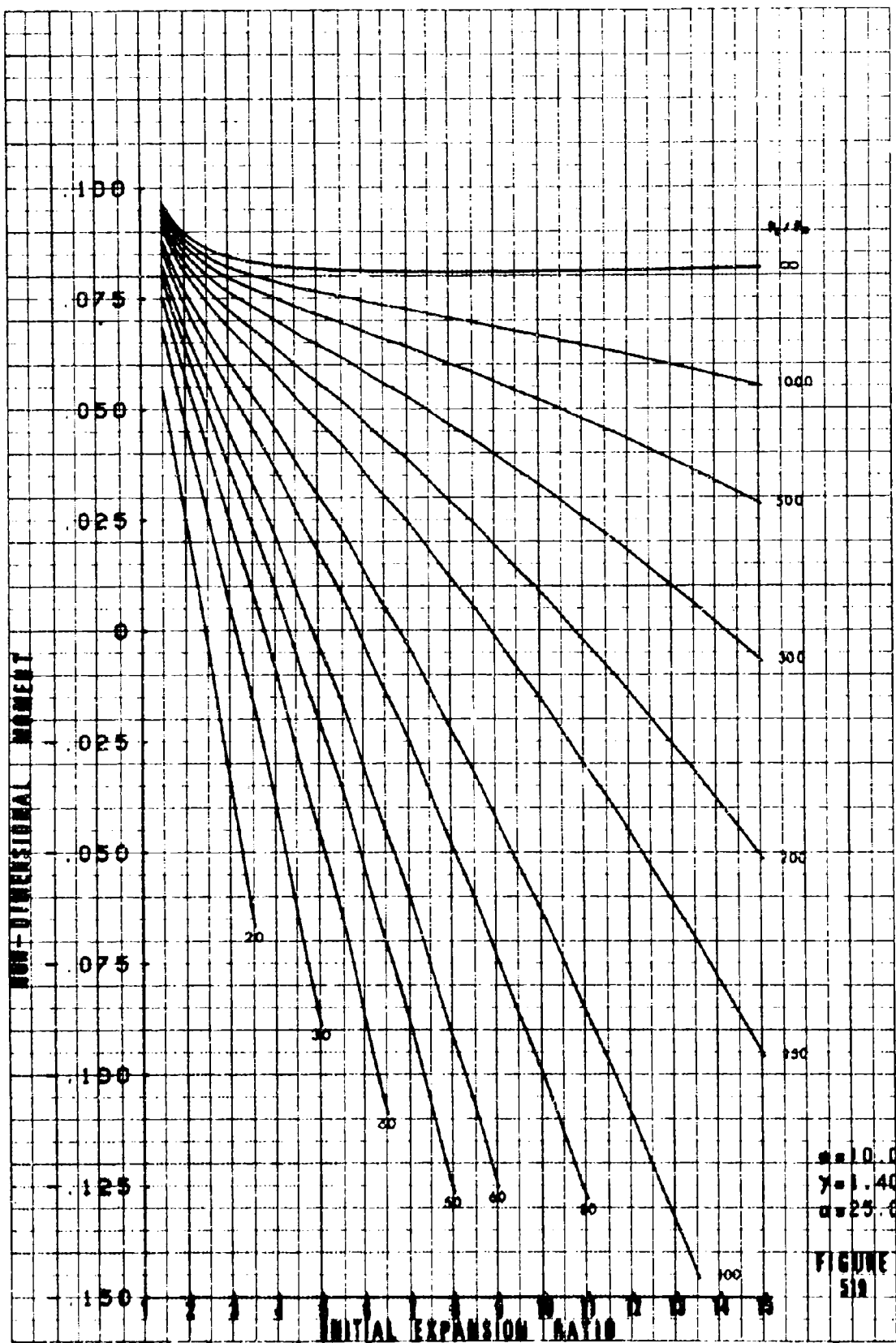


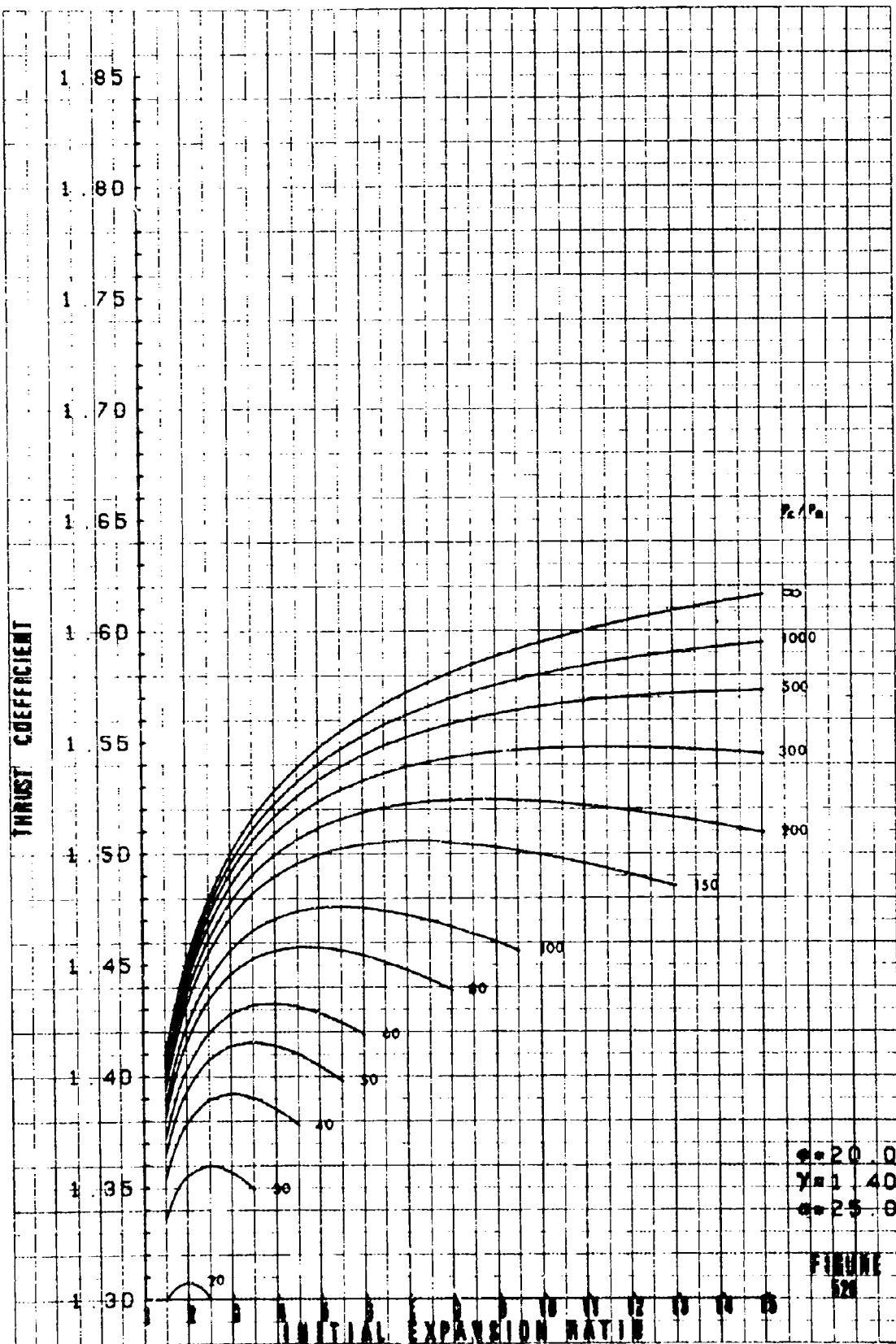
FIGURE
511

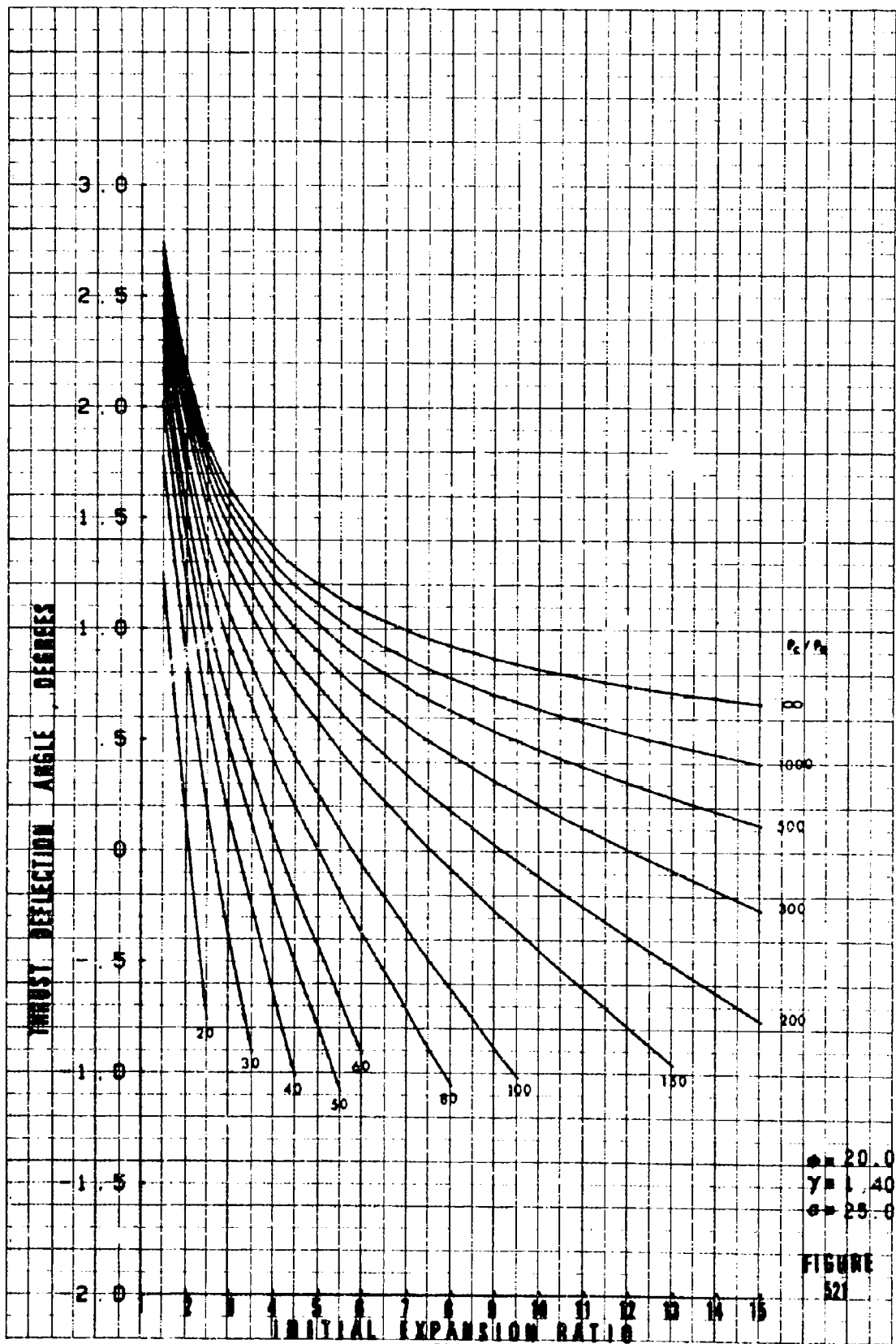


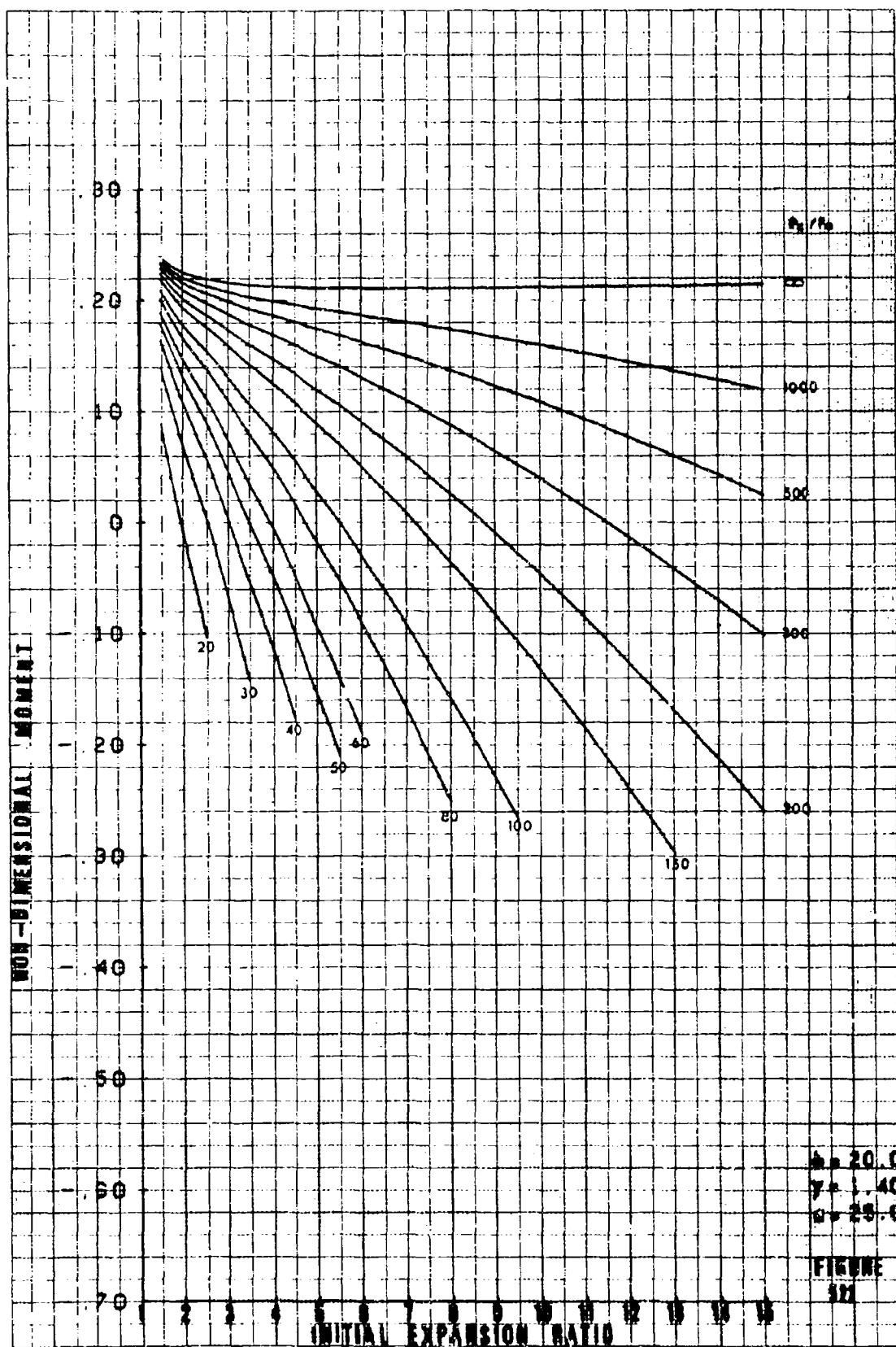
$\phi = 10.0$
 $\gamma = 1.40$
 $\sigma = 25.0$

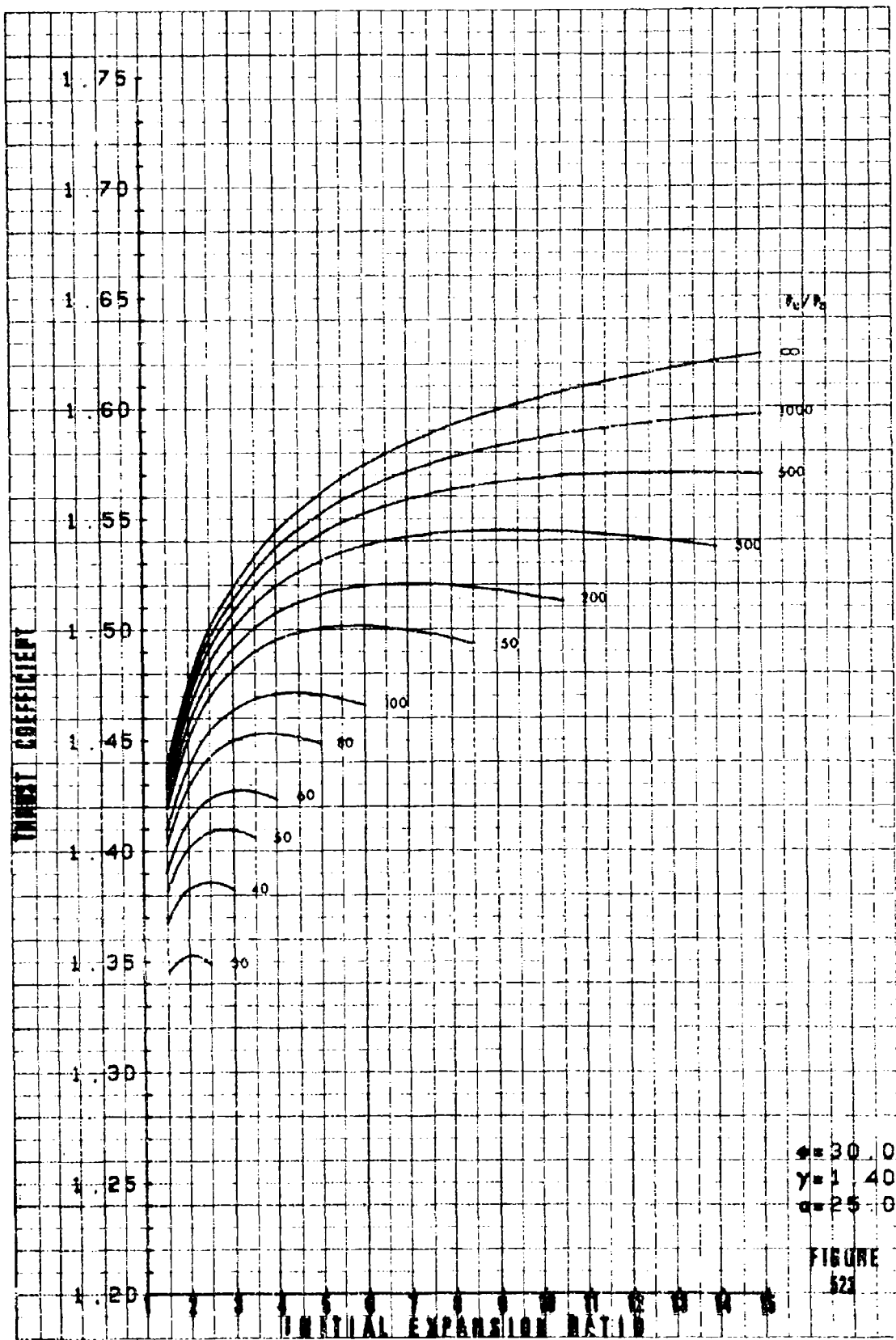
FIGURE 518





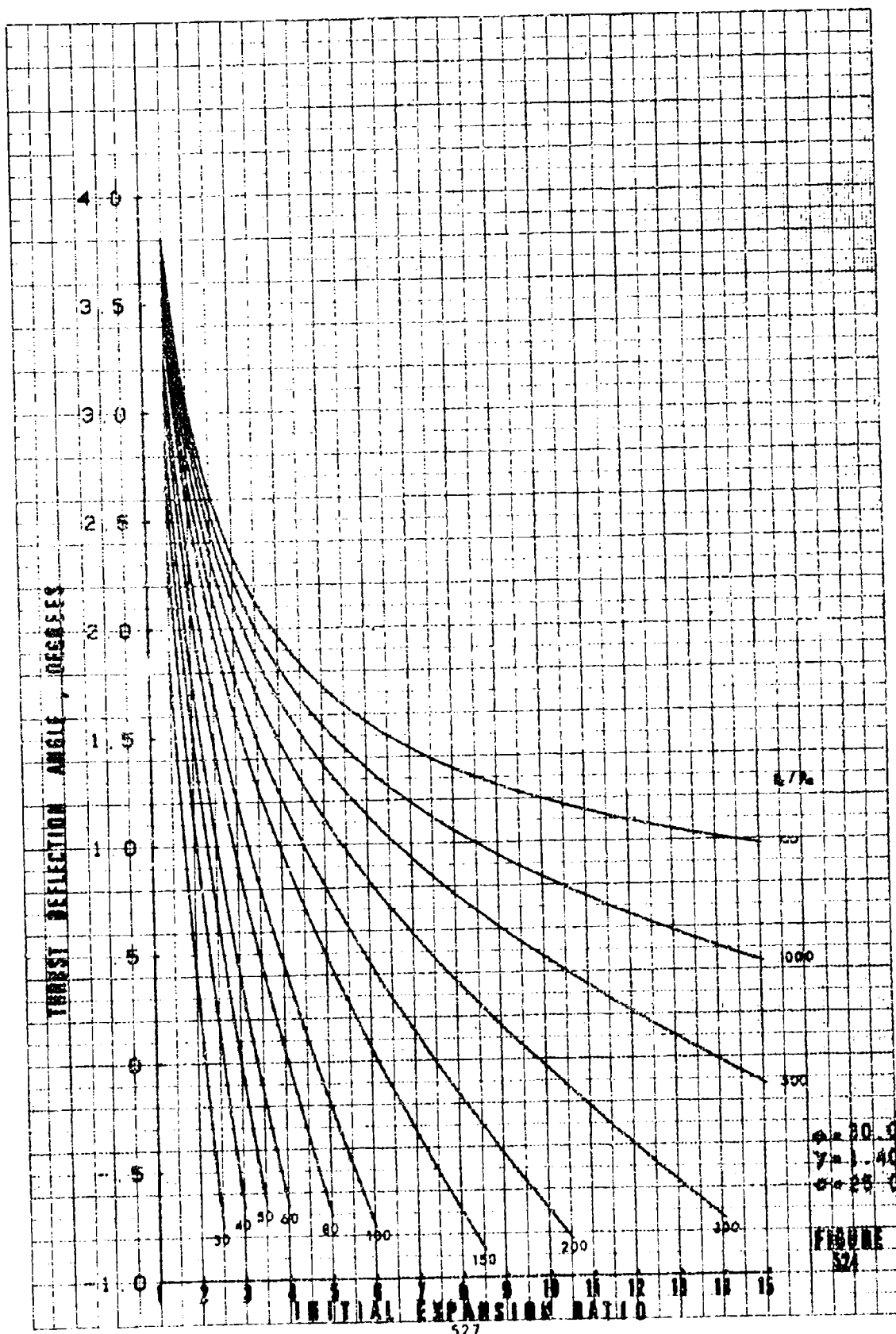


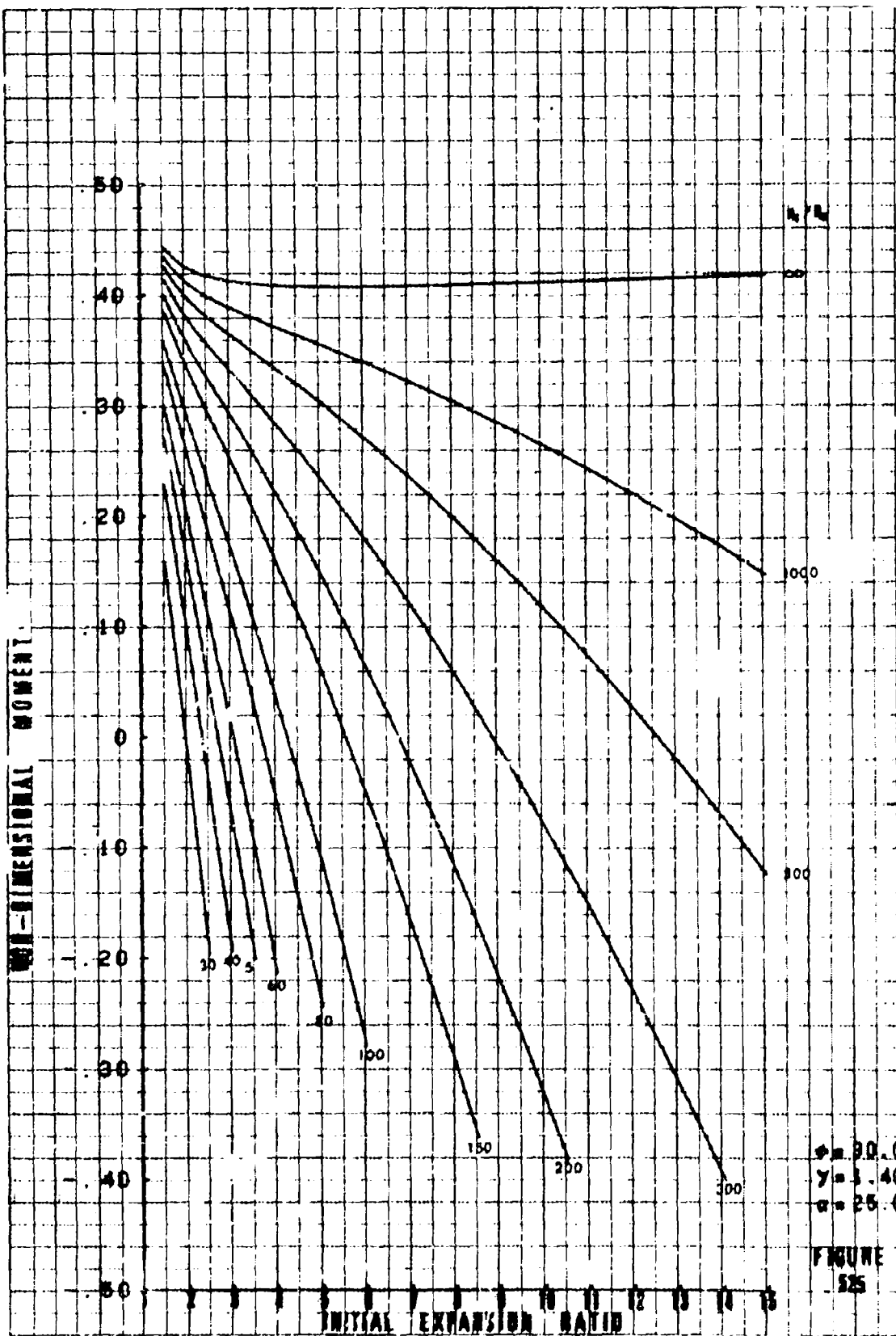


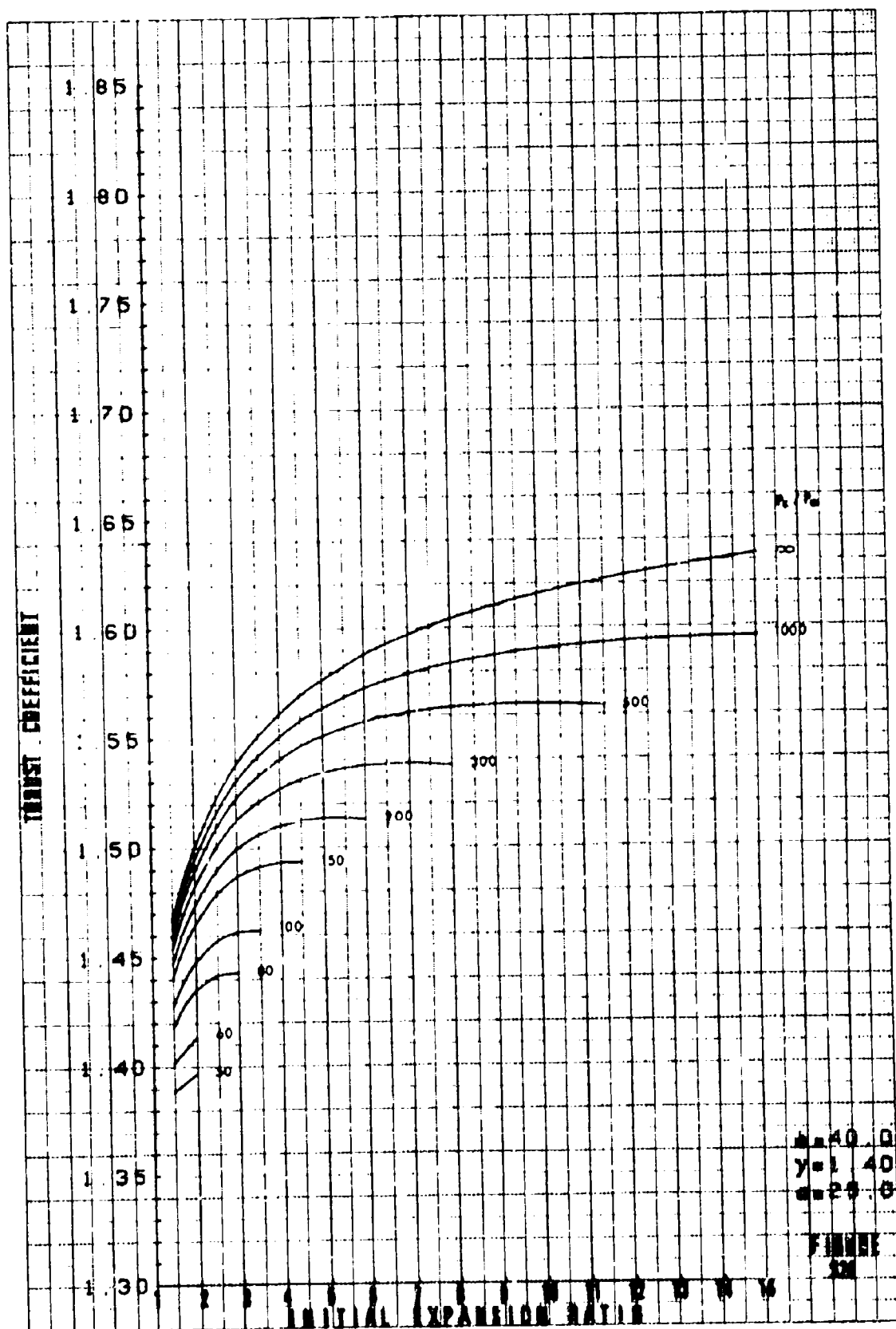


$\gamma = 1.40$
 $\alpha = 25.0$

FIGURE 523

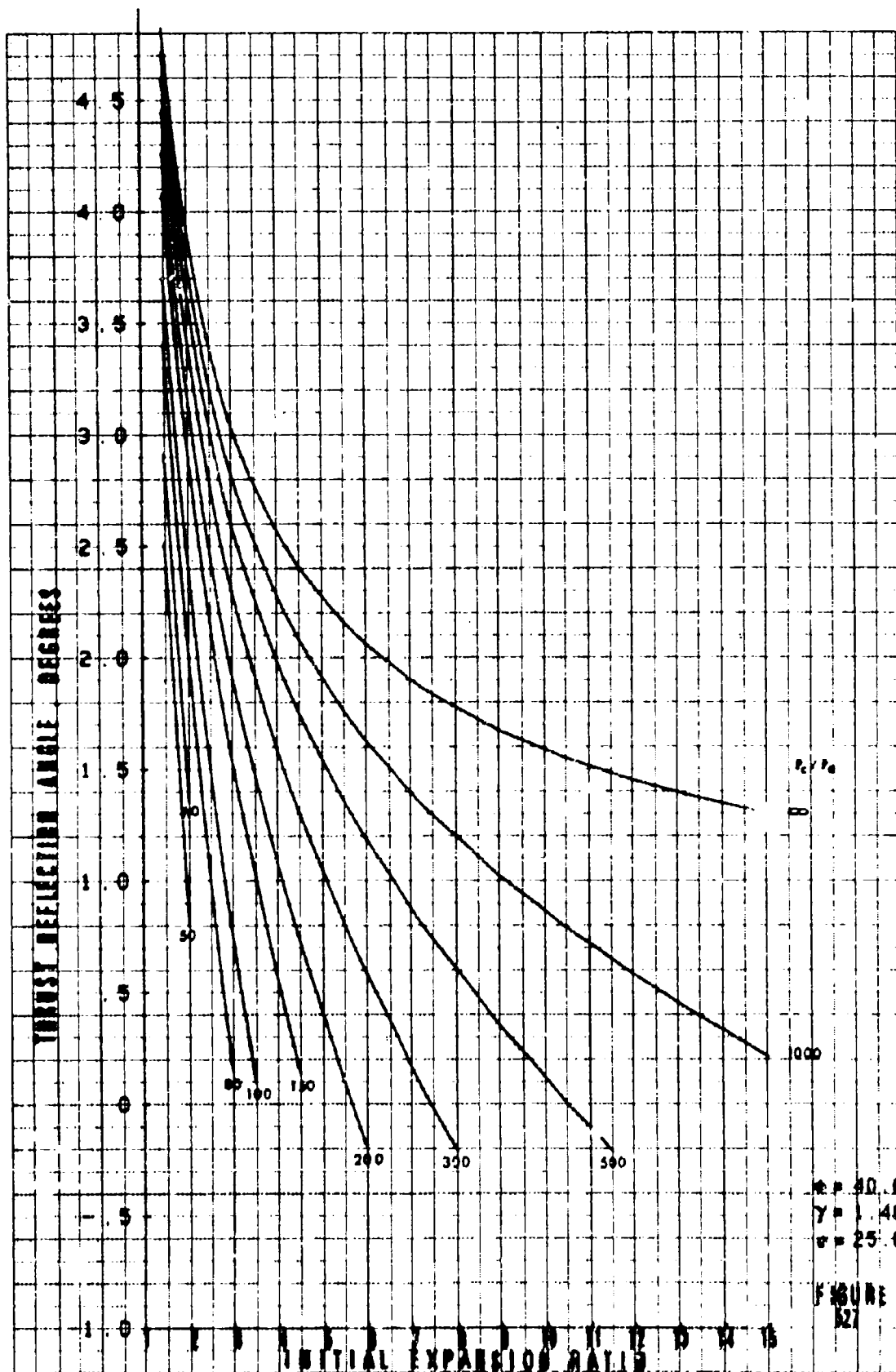






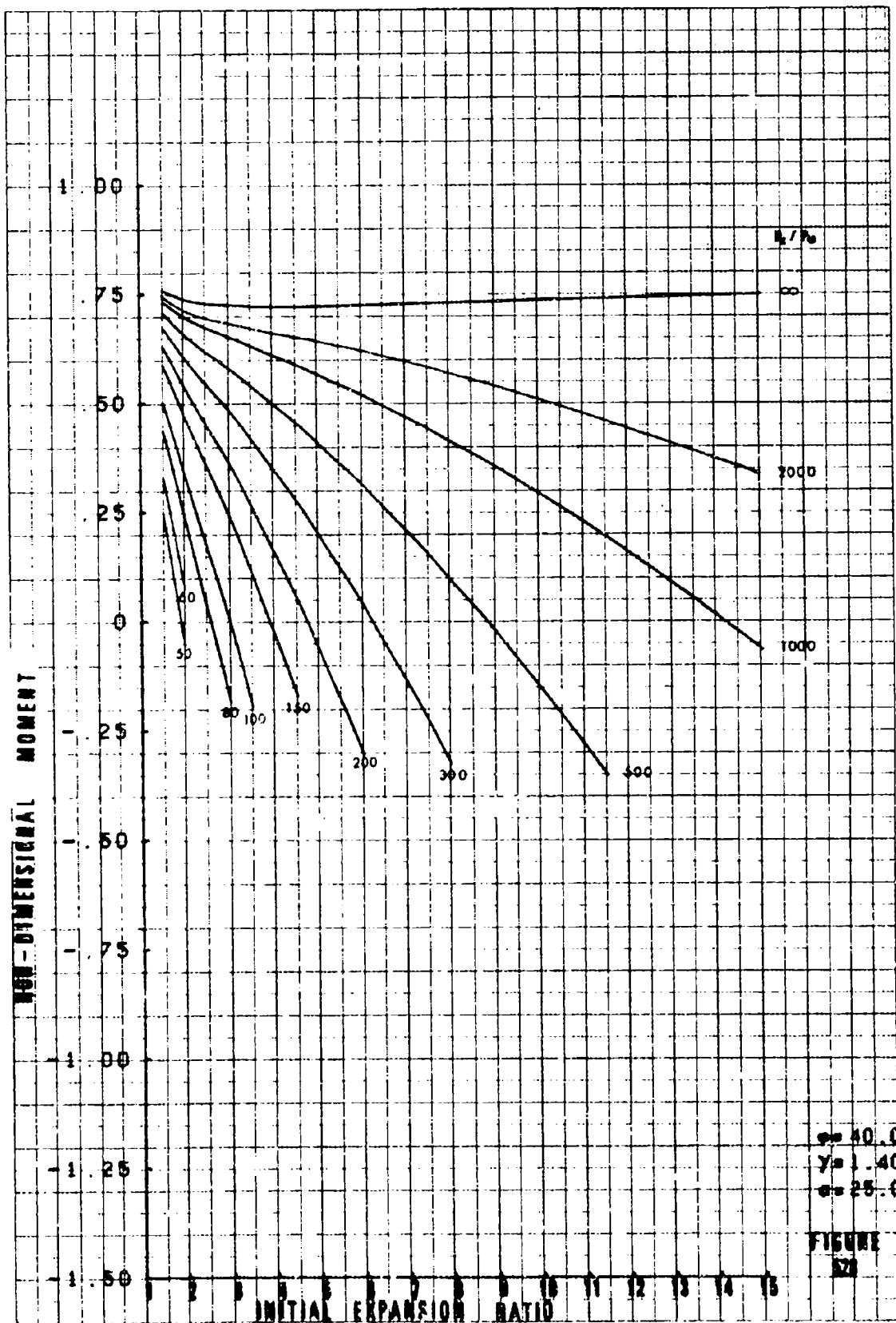
$M = 40.0$
 $\gamma = 1.40$
 $g = 29.8$

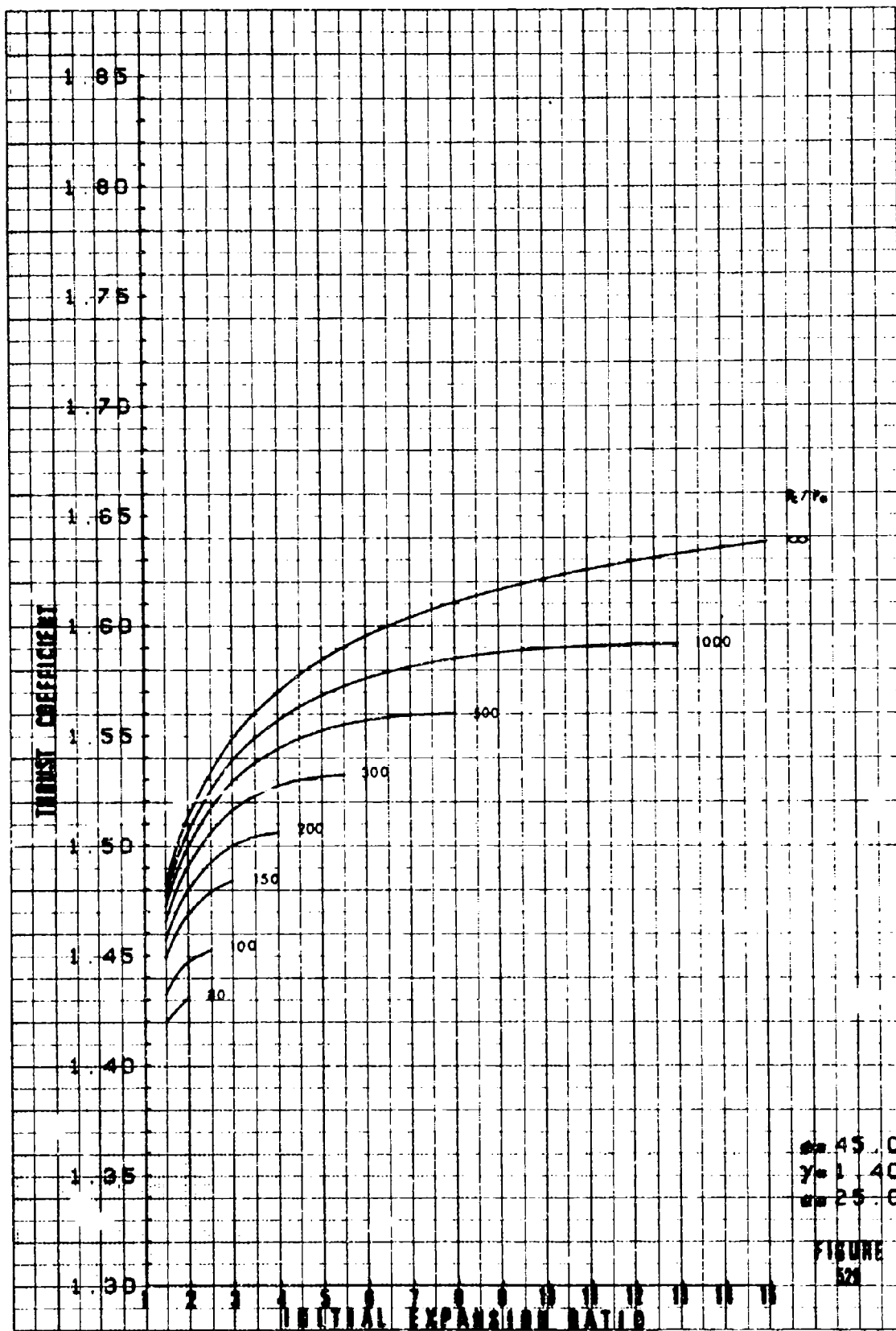
FIGURE
 32

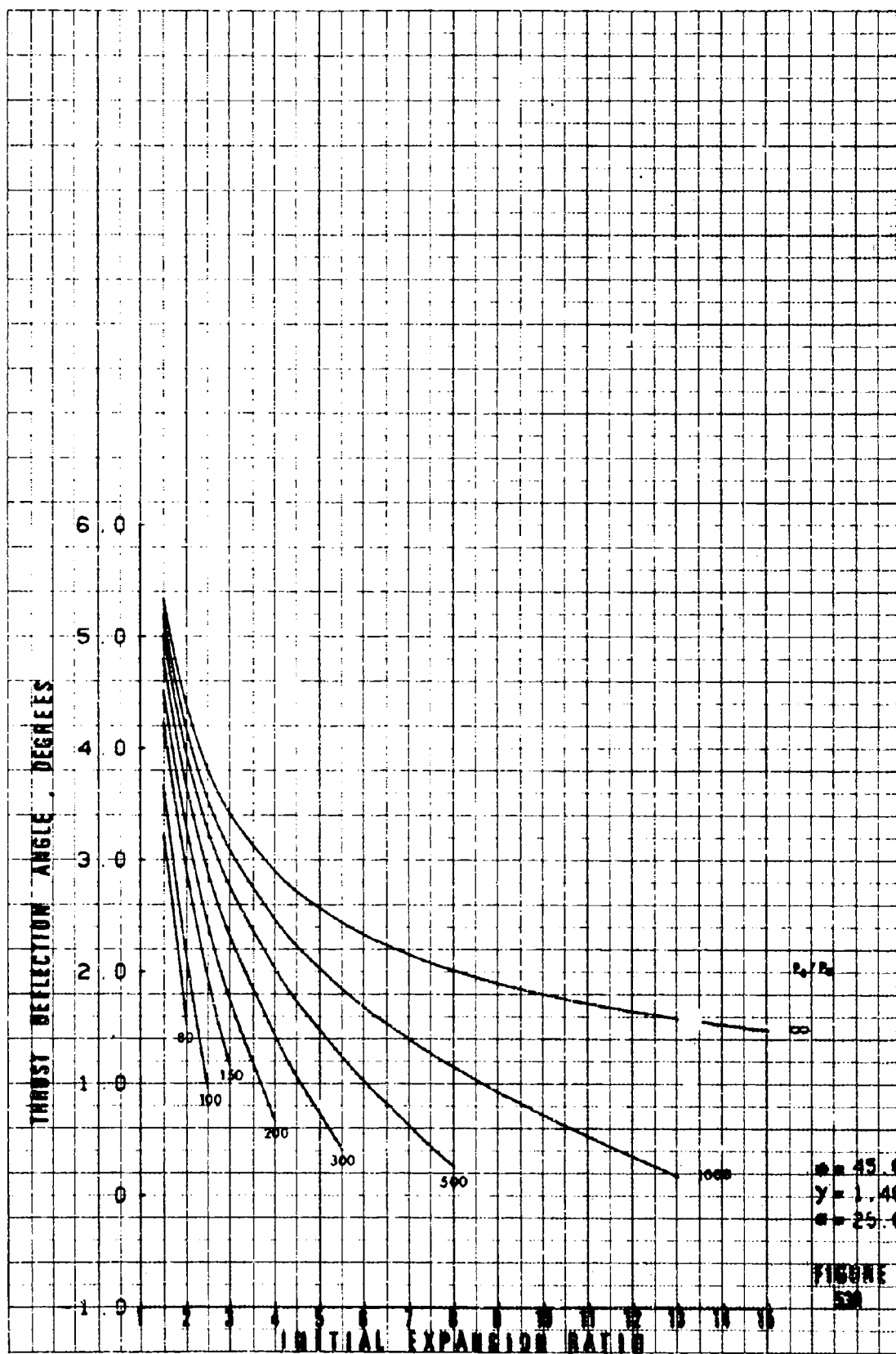


$\mu = 30.0$
 $\gamma = 1.40$
 $\sigma = 25.6$

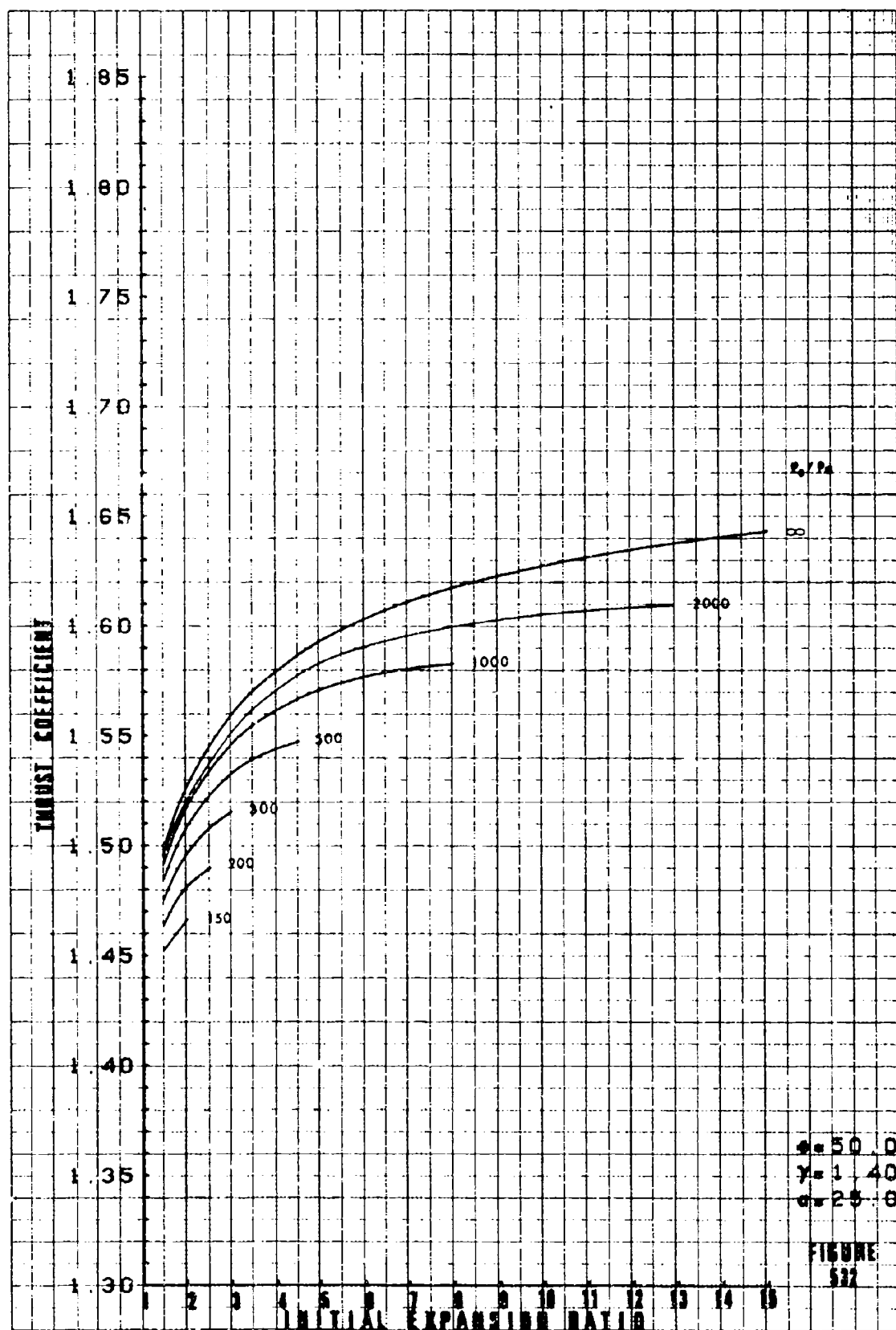
FIGURE 52











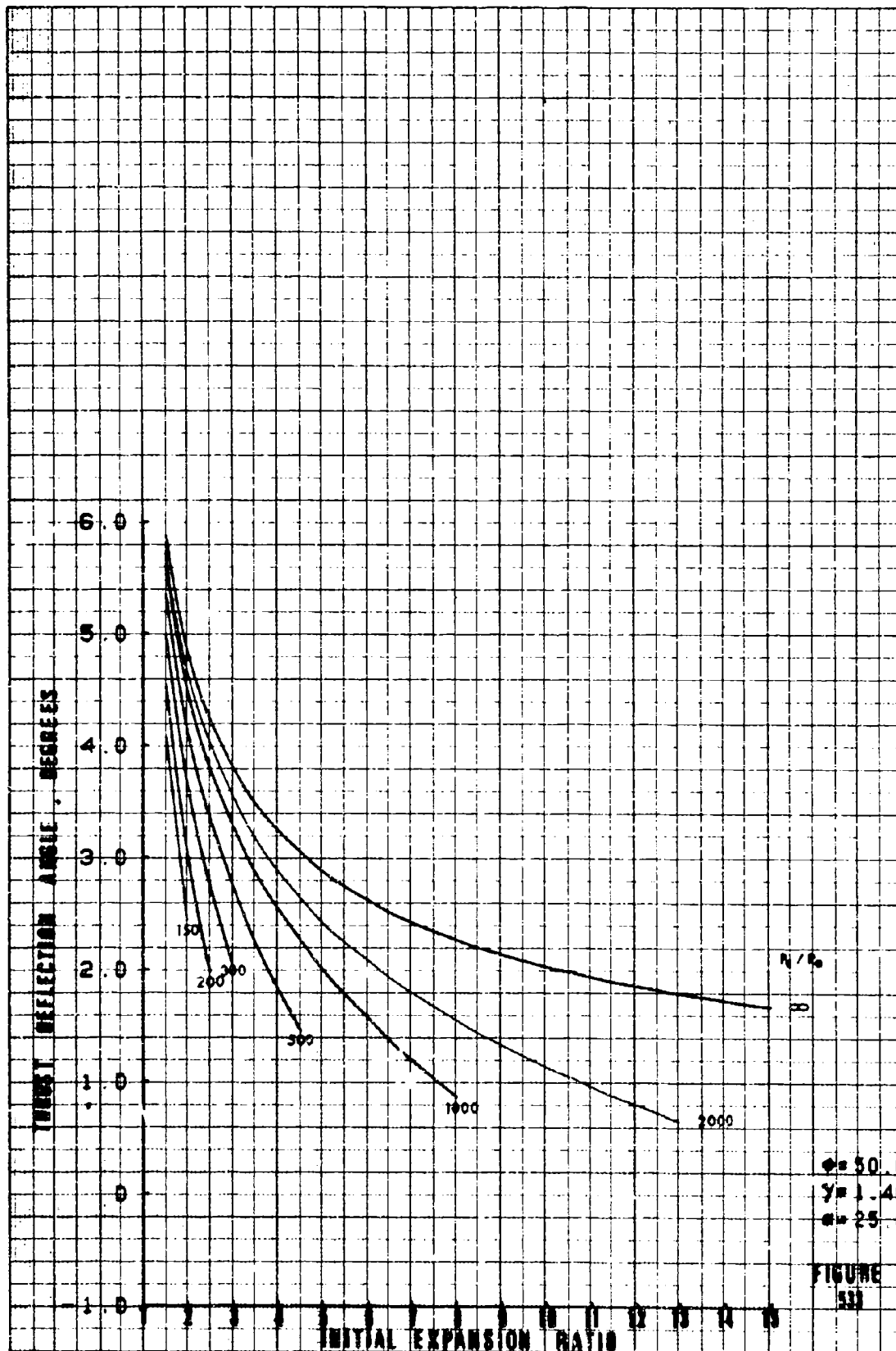


FIGURE
533

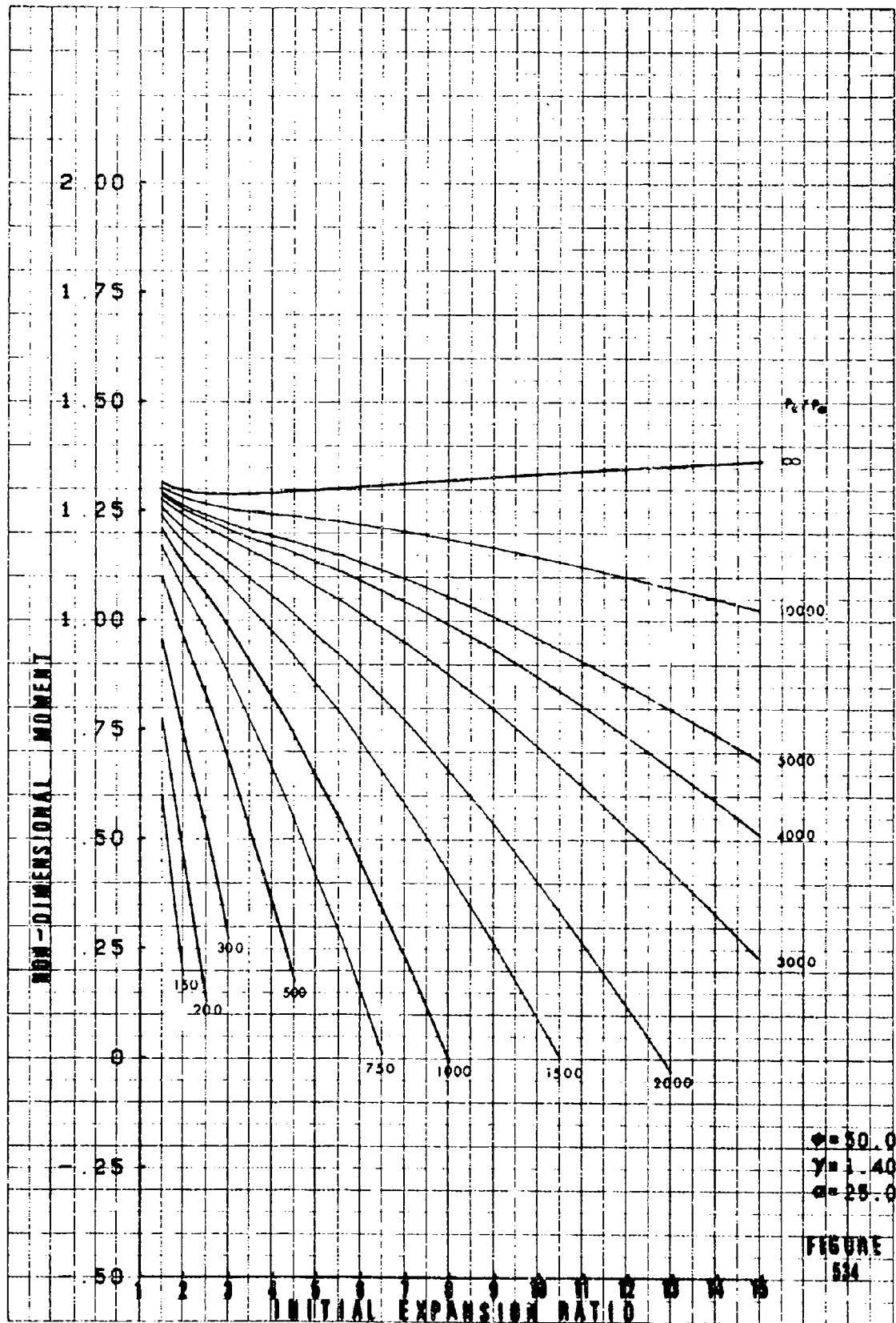
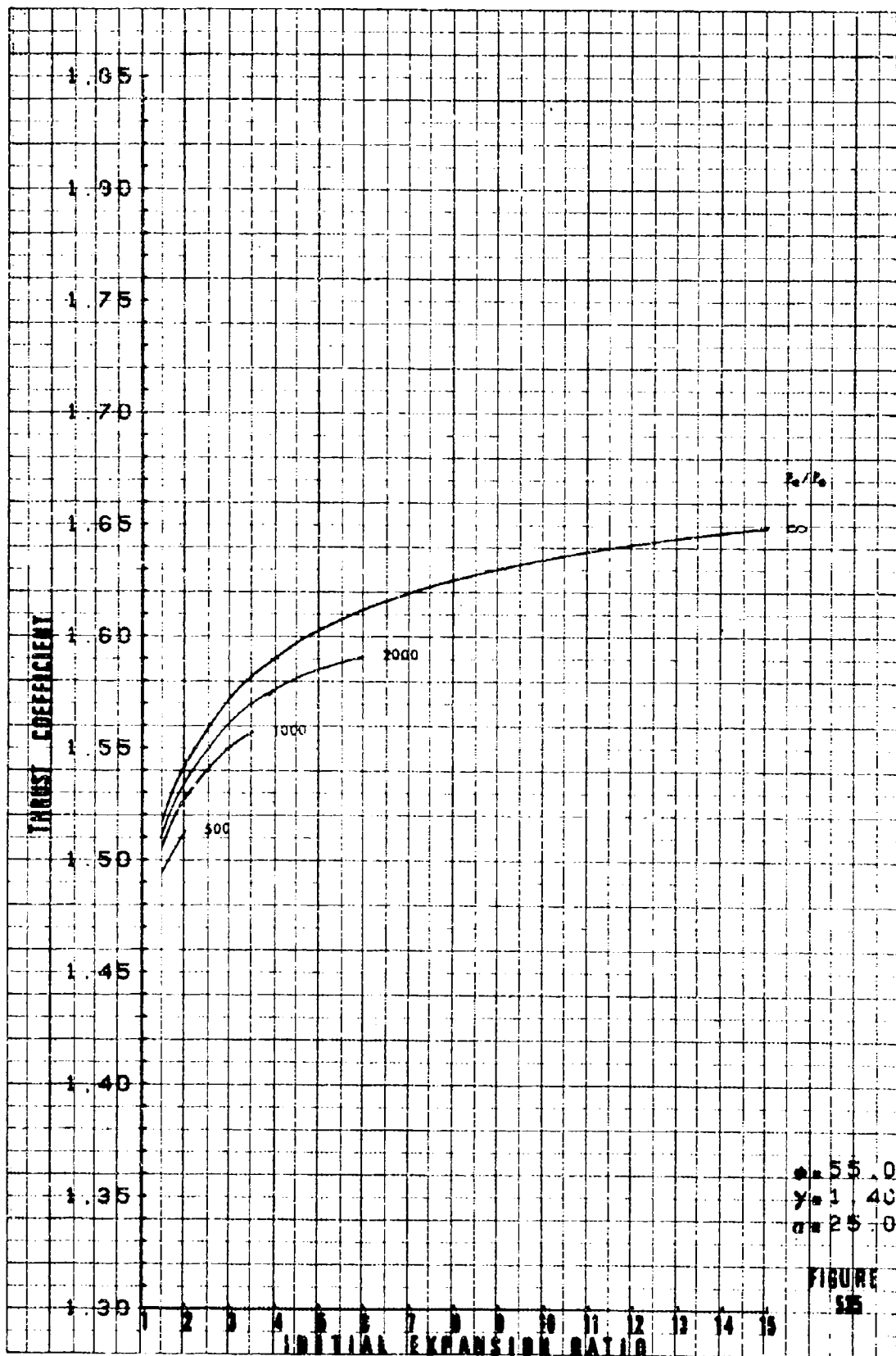
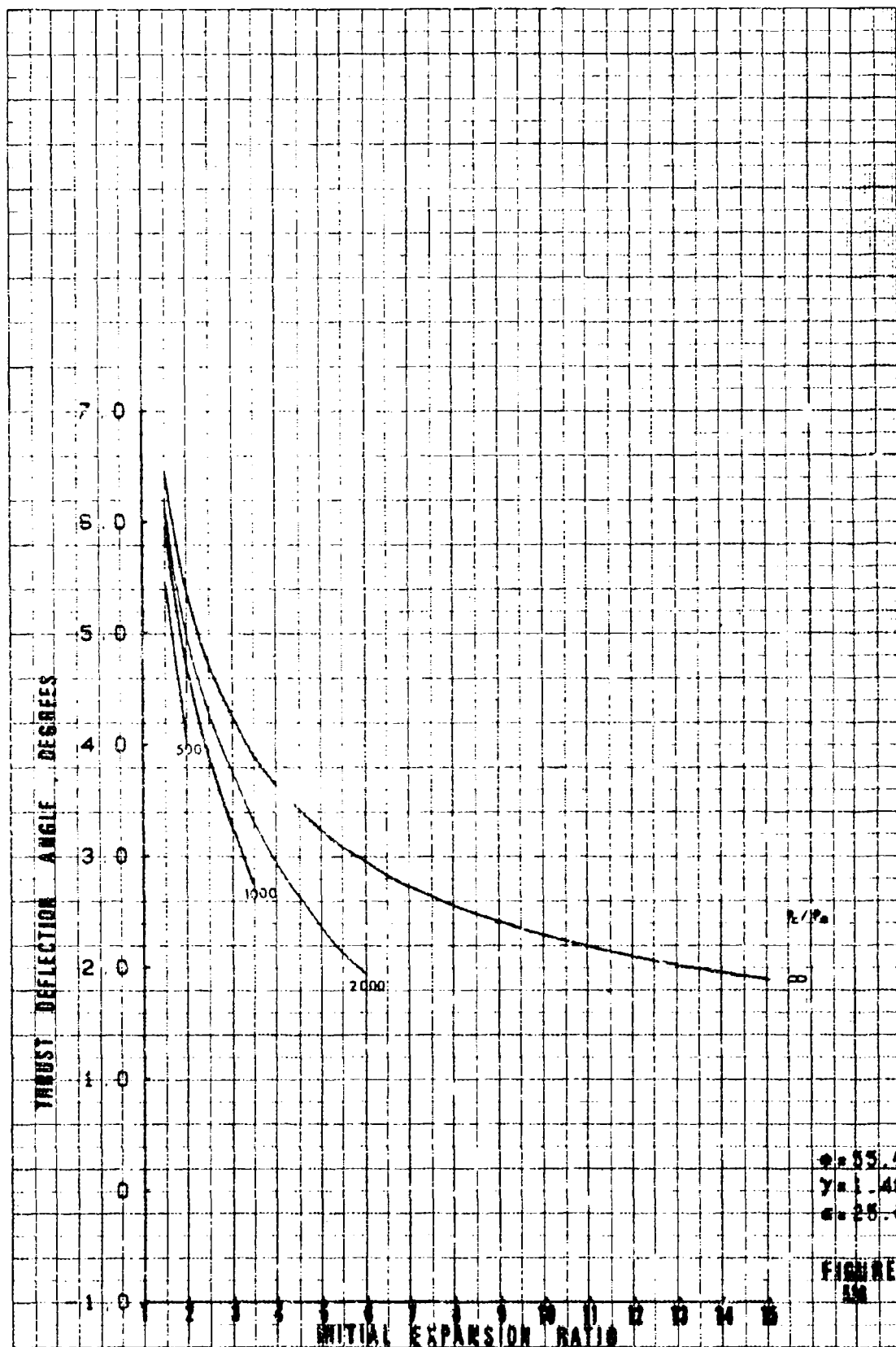
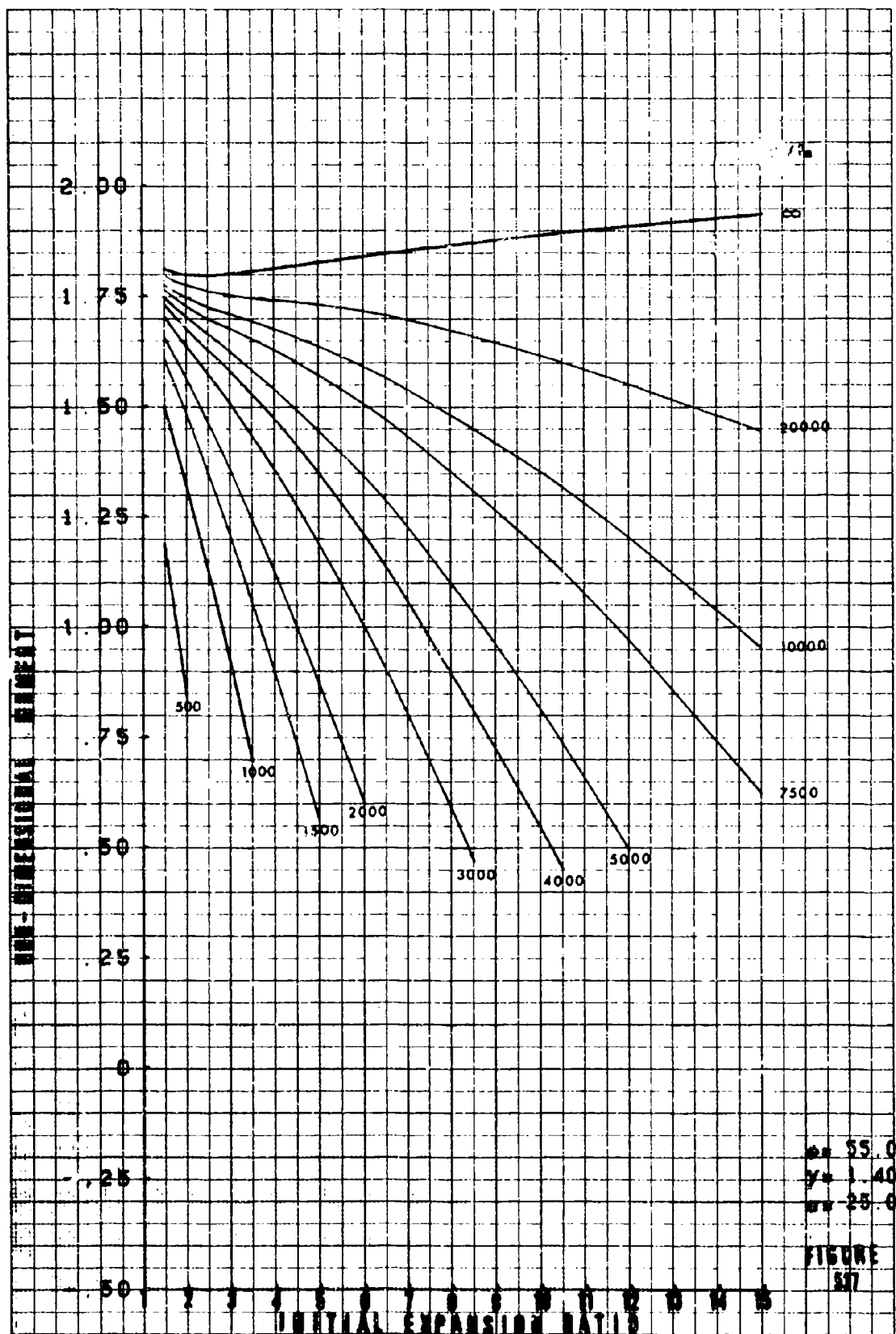
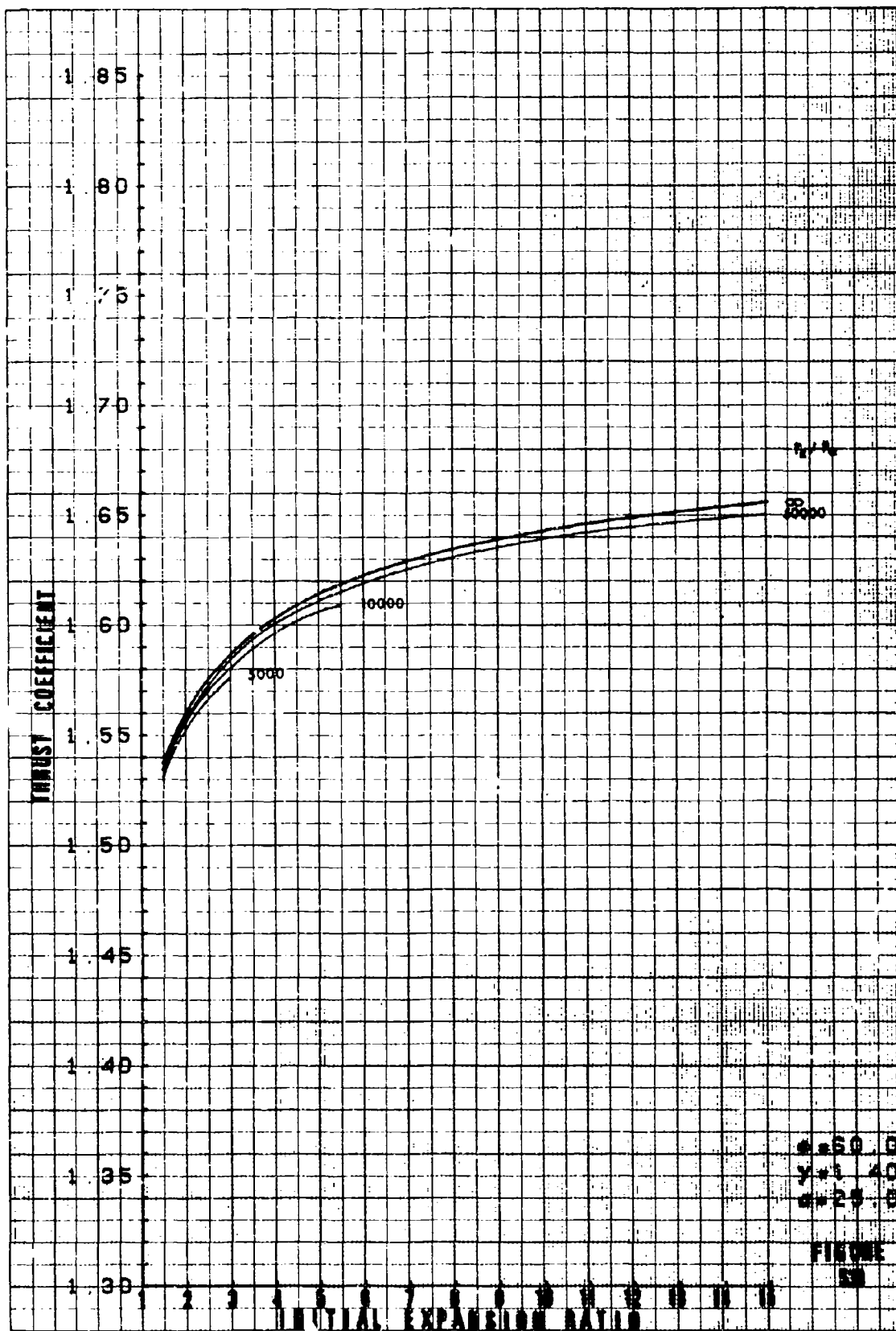


FIGURE 534









FIGURE

50

